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## NATIONAL A .... "NAUTICS AND SPACE ADMINISTRATION



(NASA-CF-147615) HEAT TRANSFER TEST OF AN O.006-SCALE THIN-SKIN THERMOCOUPLE SPACE SHUTF'S HODEL (50-0, 41-T) IN THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNMEL AT MACH 5.3 (IH28), VOLUME 1

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SPACE SHUTTLE

AEROTHERMODYNAMIC DATA REPORT

JOHNSON SPACE CENTER DA

HOUSTON, TEXAS

August, 1976

### DMS-DR-2180 NASA CR-147,615

#### VOLUME 1 OF 2

HEAT TRANSFER TEST OF AN 0.006-SCALE THIN-SKIN THERMOCOUPLE SPACE SHUTTLE MODEL (50-0, 41-T) IN THE NASA-AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC

WIND TUNNEL AT MACH 5.3 (IH28).

#### Ъy

J. W. Cummings/T. F. Foster Shuttle Aerosciences Rockwell International Space Division W. K. Lockman NASA-Ames Research Center

Prepared Under Contract Number NAS9-13247

by

Data Management Services Chrysler Corporation Space Division New Orleans, La. 70189

### for

#### Engineering Analysis Division

Johnson Space Center National Aeronautics and Space Administration Houston, Texas

#### WIND TUNNEL TEST SPECIFICS:

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Test Number:	ARC 3.5-195
NASA Series Number:	IH28
Model Number:	50-0, 41-T
Test Dates:	May 17 through May 24, 1974
Occupancy Hours:	88

### FACILITY COORDINATOR:

Joseph G. Marvin Mail Stop N-229-1 NASA-Ames Research Center Moffett Field, CA. 94035

### **PROJECT ENGINEERS:**

J. W. Cummings	T. F. Foster	W. K. Lockman
Rockwell International	Rockwell International	Mail Stop N-229-1
Space Division	Space Division	NASA-Ames Research Center
12214 Lakewood Blvd.	12214 Lakewood Blvd.	Moffett Field, CA. 94035
Mail Stop AD38	Mail Stop AD38	
Downey, CA. 90241	Downey, CA. 90241	Phone: (415) 965-6755

Phone: (213) 922-4600 Phone: (213) 922-4600

### DATA MANAGEMENT SERVICES:

Prepared by: Liaison--D. A. Sarver Operations--R. B. Lowe

Reviewed by: D. E. Poucher

Approved: J. L. Clynn, Manager Data Operations

Concurrence:

N. D. Kemp, Manager Data Management Services

Chrysler Corporation Space Division assumes no responsibility for the lata presented other than display characteristics.

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HEAT TRANSFER TEST OF AN 0.006-SCALE THIN-SKIN THERMOCOUPLE SPACE SHUTTLE MODEL (50-0, 41-T) IN THE NASA-AMES RÉSEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL AT MACH 5.3 (IH28)

by

J. W. Cummings/T. F. Foster Shuttle Aerosciences Rockwell International Space Division W. K. Lockman NASA-Ames Research Center

#### ABSTRACT

This report presents data obtained from a heat transfer test conducted on an 0.006-scale Space Shuttle Orbiter and External Tank in the NASA-Ames Research Center 3.5-foot Hypersonic Wind Tunnel. The purpose of this test was to obtain data under simulated return-to-launch-site abort conditions. Configurations tested were integrated orbiter and external tank, orbiter alone, and external tank alone at angles of attack of 0,  $\pm 30$ ,  $\pm 60$ ,  $\pm 90$ , and  $\pm 120$  degrees.

Runs were conducted at Mach numbers of 5.2 and 5.3 for Reynolds numbers of 1.0 x  $10^6$  and 4.0 x  $10^6$  per foot, respectively. Heat transfer data were obtained from 75 orbiter and 75 external tank iron-constantan thermocouples.

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This report consists of 2 volumes. Volume 1 contains Figures 4 + 15; whereas, Volume 2 contains Figures 16 + 27 and the Tabulated Source Data.

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τ <b>Λ</b>	SCHEDULE OF COEFFICIENTS PLOTTED	(A) H	(A) R	(B) F	(C) H	E (C)	<b>(</b> 0)	(E) B	(E) H	(F)	(E)	(E)	(4)
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	CONDITIONS VARYING	HAW/HT, ALPHA	HAW/HT,ALPHA,RN/L,MACH	ALPHA	HAW/HT, 2Y/B, ALPHA	HAW/HT, 2Y/B, ALPHA, RN/L, BETA, MACH	2Y/B, ALPHA	HAW/HT,2Y/B,ALPHA,X/C	HAW/HT,2Y/B,ALPHA,RN/L BETA, X/C, MACH	ZY/B, ALPHA, X/C	HAW/HT, Z, ALPHA	HAW/HT, Z, ALPHA RN/L, BETA, MACH	Z, ALPHA
Continued	EDULE OF FFICIENTS LOTTED	(3)	(c)	<b>(</b> 0 <b>)</b>	(9)	(9)	(H)	(1)	(I)	(r)	(9)	(9)	(H)
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### SCHEDULE OF COEFFICIENTS PLOTTED:

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(A)	H/HREF	versus	X/L
	H/HREF	versus	Phi
(B)	HI/HU	versus	X/L
	HI/HU	versus	PHI
(C)	H/HREF	versus	X/L
(D)	HI/HU	versus	X/L
(E)	H/HREF	vėrsus	X/L
	H/HREF	versus	Z
(F)	HI/HU	versus	X/L
	HI/HU	versus	Z
(G)	H/HREF	versus	x/c
(H)	HI/HU	versus	x/c
(1)	H/HREF	versus	X/C
	H/HREF	versus	2Y/B

(J) HI/HU versus X/C HI/HU versus 2Y/B

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### NOMENCLATURE

SYMBOL	PLOT SYMBOL	DEFINITION
ъ		thickness of model skin, in.
B	BREF	span length, in.
C		specific heat of model skin material, $BTU/lb_m - {}^{O}R$
c		chord length, in.
C <sub>0</sub> , C <u>1</u> , C2		constants in curve fit for C over model wall temperature range
ср		specific heat of air stream (perfect gas value), BTU/lbm- <sup>O</sup> R
CHAN	CHAN	Recording-system channel
Haw	HAW	adiabatic wall enthalpy, BTU/15m
Ht	HT	free-stream total enthalpy, BTU/1bm
	НО	average of free-stream total enthalpy values of all tunnel runs incorporated into an aero data- set, BTU/lb <sub>m</sub>
Hwi	WH	enthalpy based on model wall temperature for given T/C location at initial time, BTU/lbm
h	Ħ	heat-transfer coefficient at model wall for given T/C location
hs	hs, href	stagnation-point heat-transfer coefficient for reference sphere
h/h <sub>s</sub>	h/HS, H/HREF	ratio of model heat-transfer coefficient to heat-transfer coefficient of reference sphere for $H_{\rm RW}/H_{\rm t}$ = X.XXX
IML		inner mold line
L	lref, length	model reference length, in.or ft.
Moo	MACH	free-stream Mach number
Hw		enthalpy based on model wall temperature, BTU/1bm

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## NOMENCLATURE (Continued)

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SYMBOL	SYMBOL	DEFINITION
Pt	PT	free-stream total pressure, psia
	20	average of free-stream total pressure valuer of all tunnel runs incorporated into an aero dataset, psia
• 9 <u>1</u>	QDOT, Q	heat-transfer rate at model wall for given T/C location at initial time, BTU/ft <sup>2</sup> -sec
ġ s	QS, QREF	stagnation-point heat-transfer rate for ref- erence sphere at initial time, BTU/ft <sup>2</sup> -sec
R <sub>8</sub>	RS	reference sphere radius at model scale equiva- lent to 0.305 m (1 ft) for full-scale vehicle
Re∞/ft	RE/FT	free-stream Reynolds number per foot
	RN/L	average of free-stream Reynolds number values (per foot) of all tunnel runs incorporated into an aero dataset
$\operatorname{Re}_{\infty}, L$	REL	free-stream Reynolds number based on model reference length, L
	s/r	body wetted running length
St	ST	Stanton number based on free-stream flow conditions and the model heat-transfer coefficient for H <sub>aw</sub> /H <sub>t</sub> = X.XXX
Т		terperature, <sup>O</sup> R
Tt	IT	free-stream total temperature, <sup>O</sup> R
	TO	average of free-stream total temperature values of all tunnel runs incorporated into an aero dataset, R
Tw i	Tw	model wall temperature for given T/C loca- tion at initial time. <sup>O</sup> R

## NOMENCLATURE (Continued)

SYMBOL	PLOT SYMBOL	DEFINITION .
т/с	т/с	thermocouple
t		time, sec
ti	TIME	initial time (before model insertion into flow) extrapolated from $f(T_W)$ vs. time, sec
u,V		velocity, ft/sec
W		density of model skin material $lb_m/ft^3$
x		axial distance measured from nose, in.
	x/c	chordwise location, fraction of local chord
	X/L	longitudinal location, fraction of body length
Y		spanwise distance from centerline, in.
2у/в	2Y/B	spanwise location, fraction of semi-span
Z	Z	water plane distance, in.
	z/bv	spanwise location on vertical tail, fraction of expessed span
α	Alpha	angle of attack, degrees
β	BETA	angle of sideslip, degrees
μ		viscosity of air, lb-sec/ft <sup>2</sup>
ρ		density of air, 1bm/ft3
θ	THETA.	external tank angular surface coordinate, measured clockwise looking forward. O degrees at bottom centerline, degrees
ø	PHI	orbiter ängular surface coordinate, measured clockwise looking forward. O degrees at bottom centerline. degrees

### NOMENCLATURE (Concluded)

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SYMBOL	PLOT SYMBOL	DEFINITION
W.P.		water plane, height measured along Z axis, in.
B.L.	BP	butt plane, distance from orbiter centerline in the outboard direction, in.
	HI/HU	ratio of interference to undisturbed heat transfer coefficients
	ZMRP	moment referenc: point on Z axis
	YMRP	moment reference point on Y axis
	XMRP	moment reference point on X axis
	SREF	reference length or wing mean aerodynamic chord; ft.
SUBSCRIPT	<u>'S</u>	
aw		adiabatic vall
ť		initial value before model insertion
0		Orbiter
PG		perfect gas (calorically and thermally perfect gas)
8		reference sphere
t		free-stream total condition
т		tank
v		vertical tail
W		wall
<b>œ</b>		free-stream

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#### CONFIGURATIONS INVESTIGATED

The model (Orbiter and External Tank) tested was a 0.006-scale representation of the Rockwell International Space Shuttle Vehicle. The Orbiter and External Tank are defined by Rockwell lines SS-H-O11:14 and SS-H-O1415.

The Orbiter and Tank were initially built by Grumman Aircraft, Bethpage, New York, but the Orbiter was modified with additional thermocouples added to the upper surface of the left wing, vertical tail, and OMS pod. Modifications of both Orbiter and External Tank stings were accomplished to carry increased loading within the high angle of attack range.

The Orbiter was a full span (cast stainless steel) model with thinskin inserts. Thin-skin stainless steel (17-4PH) inserts were located on the underside region. left-hand wing (top and bottom), windshield area, left fuselage side, OMS pod, and vertical tail. These inserts were instrumented with 89 iron-constantan thermocouples of which only 75 were used during this test. The model was built with all control surfaces in the  $0^{\circ}$  deflection condition.

The External Tank was constructed of thin-skin (15-5PH) stainless steel. The Tank was instrumented with 111 iron-constantan thermocouples, of which only 75 were used.

The Orbiter and External Tank were designed so either could be tested alone or in the second stage configuration.

# CONFIGURATIONS investigated\_(Concluded)

The following configuration components were tested:

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Notation	Description
<sup>B</sup> 22	Fuselage (1-147B Lines)
°7	Canopy
<b>F</b> 5	Body Flap
M <sub>14</sub>	oms Pods
v <sub>7</sub>	Vertical Tail
₩ııı	Wing
T <sub>8</sub>	External Tank (-139 Lines)

### MODEL INSTRUMENTATION

The Orbiter and External Tank were instrumented with 200 ironconstantan\_thermocouples, but only 150 were used for this test. All thermocouples were spotwelded to thin-skin (nominal skin thickness of 0.030 in.) stainless steel inserts and the leads were clamped in bundles within the model. The exact T/C locations for the Orbiter and External Tank are presented in Tables IV and V, respectively, and illustrated in Figures 2a and 2b, respectively. The T/C leads were 50 feet long and fitted with Cannon Plug connectors.

#### TEST FACILITY DESCRIPTION

The NASA-Ames 3.5-foot Hypersonic Wind Tunnel is a closed-circuit, blowdown-type tunnel capable of operating at nominal Mach numbers of 5, 7, and 10 at pressures to 1800 psia and temperatures of  $3400^{\circ}$ R for run times to four minutes. The major components of the facility include a gas storage system where the test gas is stored at 3000 psi, a storage heater filled with aluminum-oxide pebbles capable of heating the test gas to  $3400^{\circ}$ R, axisymmetric contoured nozzles with exit diameters of 42 inches for generating the desired Mach number, and a 900,000 ft<sup>3</sup> vacuum storage system which operates to pressures of 0.3 psia. The test section itself is an open-jet type enclosed within a chamber approximately 12-feet in diameter and 40-feet in length, arranged transversally to the flow direction.

A model support system is provided that can pitch models through an angle-of-attack range of -20 to +20 degrees, in a vertical plane, about a fixed point of rotation on the tunnel centerline. This rotation point is adjustable from 1 to 5 feet from the nozzle exit plane. The model normally is out of the test stream (strut centerline 37 inches from tunnel centerline) until the tunnel test conditions are established after which it is inserted. Insertion time is adjustable to as little as  $\frac{1}{2}$  second and models may be inserted at any strut angle.

A high-speed, analog-to-digital data acquisition system is used to record test data on magnetic tape. The present system is equipped to measure and record the outputs from 80 transducers in addition to 20 channels of tunnel parameters.

#### TEST PROCEDURE

Heat Transfer Data were obtained by measuring the temperature rise over a period of time from a total of 150 iron-constantan thermocouples. The model was injected into the flow in approximately 1 second and held on tunnel centerline for approximately 1 second. Temperature measurements and tunnel conditions were recorded on magnetic tape at 0.07second intervals by the data acquisition system from the start of model injection to the start of model retraction.

A maximum of 75 thermocouples could be recorded for any given run. The thermocouple leads were routed from the model through the tunnel model-injection mechanism, and connected to a junction box which was wired directly to a thermocouple reference-temperature (150°F) box. The junction box connectors were wrapped with asbestos for heat protection from the tunnel test-chamber ambient conditions (no free-stream flow on box). Thermocouple changes were accomplished by changing 5 Cannon Plugs containing 15 thermocouples each. Prior to testing, a thermocouple heat-response check, through the data-acquisition system, was performed on all thermocouples to assure proper hook-up, polarity and response.

Prior to each run with model attitude changes, the model was leveled in pitch and roll by means of leveling blocks which attach to the sting assembly of the Orbiter/External Tank. When leveling the models, an inclinometer was placed on the leveling plate. Proper roll relationships between the models were set using scribed lines on the model stings.

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### DATA REDUCTION

All test data were reduced at the NASA/Ames Research Center using the data-reduction techniques outlined below. The thermocouple data were reduced using the one-dimensional, thin-wall equation:

$$\dot{q} = WCb \frac{dT_W}{dt} = h (H_{RW} - H_W) \equiv hH_t \left( \frac{H_{RW}}{H_t} - \frac{H_W}{H_t} \right)$$
 (1)

which neglects heat-conduction losses.

Assuming that W and h are constant and

$$C = C_0 + C_1 T_V + C_2 T_V^2 \text{ for } T_V \text{ ranges}$$
(2)

the integration of equation (1) for  $t = t_i$  to t and  $T_W = T_{W_i}$ to  $T_W$  yields the linear equation:

$$f(T_W) = -\ln\left(\frac{T_{WW}^* - T_W}{T_{WW}^* - T_{W_1}}\right) - \left[\frac{C_1}{C_W^*} + \frac{C_2}{C_W^*} + \frac{T_W + T_{W_1}}{2}\right](T_W - T_{W_1})$$
$$= \frac{hc_p}{WC_{WW}^*} \quad (t - t_1) \quad (3)$$

where it is defined that:

$$\mathbf{T}_{\mathbf{A}\mathbf{V}}^{*} = \frac{\mathbf{H}_{\mathbf{B}\mathbf{V}}}{\mathbf{o}_{\mathbf{p}}} = \frac{\mathbf{H}_{\mathbf{B}\mathbf{V}}}{\mathbf{H}_{\mathbf{t}}} \stackrel{\mathbf{H}_{\mathbf{t}}}{\mathbf{e}_{\mathbf{p}}} \stackrel{\mathbf{a}}{\mathbf{e}} (\mathbf{T}_{\mathbf{B}\mathbf{V}}) \mathbf{P}\mathbf{G}$$
(4)

$$C_{av}^{i} \equiv C_{0} + C_{1} T_{av}^{i} + C_{2} T_{av}^{i}$$
 (5)

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# specific heat at adiabatic wall temperature

### DATA REDUCTION (Continued)

The form of Eq (3) is  $f(T_W) = mt + a$  where m is the slope and a is the intercept for a straight line if heat-conduction errors are negligible. Thus, deviations from a straight line can indicate heat-conduction effects.

The slope, m, of  $f(\mathfrak{T}_W)$  vs t from Eq (3) is computed by a leastsquares, straight-line fit over a finite time interval (approx. 1 sec) beginning when the model reaches uniform tunnel flow. The value of the heat-transfer coefficient, h, is then determined from:

$$h = \frac{W_{awb}^{\mu}}{c_p} m$$
 (6)

Using this value of h, the heat-transfer rate is evaluated at the initial time,  $t_1$ , when the model is isothermal at the initial wall enthalpy,  $E_{w_1}$ 

$$\hat{q} = \hat{q}_{\underline{i}} = i \left(H_{RW} - H_{W\underline{i}}\right) \equiv hH_{\underline{i}} \left(\frac{H_{RW}}{H\underline{i}} - \frac{H_{W\underline{i}}}{H\underline{i}}\right)$$
 (7)

where  $H_{\rm RW}/H_{\rm t}$  is the same value used to evaluate h. The resultant value of  $\dot{q}$  is independent of the value of  $H_{\rm RW}/H_{\rm t}$  used for both the h and  $\dot{q}$ evaluations.

The reference sphere heating is also evaluated at the initial wall enthalpy by the method of Fay and Riddell

$$\dot{\mathbf{q}}_{\mathbf{g}} = \mathbf{h}_{\mathbf{g}} \left( \mathbf{H}_{\mathbf{t}} - \mathbf{H}_{\mathbf{w}_{\mathbf{i}}} \right) \equiv \mathbf{h}_{\mathbf{g}} \quad \mathbf{H}_{\mathbf{t}} \quad \left( \mathbf{1.0} - \frac{\mathbf{H}_{\mathbf{w}_{\mathbf{i}}}}{\mathbf{H}_{\mathbf{t}}} \right)$$
(8)

The model-to-sphere ratio of heat-transfer coefficients is then determined from Eqs. (7) and (8) as

$$\frac{\mathbf{h}}{\mathbf{h}_{0}} = \frac{\mathbf{\hat{q}}_{1}}{\mathbf{\hat{q}}_{0}} \left[ \frac{\mathbf{1}_{0}\mathbf{0} - \mathbf{\hat{u}}_{1}}{\mathbf{h}_{0}} \frac{\mathbf{h}_{0}}{\mathbf{h}_{0}} / \mathbf{h}_{0} - \mathbf{h}_{0} / \mathbf{h}_{0}} \right]$$
(9)

## DATA REDUCTION (Concluded)

where  $q_i$  is constant for all values of  $H_{aw}/H_t$ . To determine  $h/h_s$  for various values of  $H_{aw}/H_t$ . the particular value of  $H_{aw}/H_t$  is substituted into Eq. (9).

The Stanton number is defined as

$$St \equiv \frac{h}{\rho u} = \frac{\dot{q_1}}{\rho u(H_{aw} - H_{w1})}$$
(10)

where for free-stream conditions,  $\rho u = \rho_{\infty} V_{\infty}$  .

The calculations of the model heating, reference sphere heating, and Reynolds number included the corrections of NACA report 1135 (Ref. 3) for calorically imperfect, thermally perfect air. Keyes' equation for viscosity (Ref. 4) was also used for the sphere heating and Reynolds number computations:

$$\mu = \frac{0.0232 \times 10^{-6} T^{0.5}}{1 + \frac{220}{T} \times 10^{-9/T}}$$
(11)

where the units for T and  $\mu$  are <sup>O</sup>R and lb-sec/ft<sup>2</sup>, respectively.

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T: 1H28 (A)	RC 3.5-195)		DATE: 6-8-74
	TEST CON	IDITIONS	
			•
MACH NUMBER	REYNOLDS NUMBER (per foot)	TOTAL PRESSURE (pounds/sq. (nch)	TOTAL TEMPERA (degrees Rankine
5.2	1.0 x 10 <sup>6</sup>	100	1500°R
5.3	4.0 x 10 <sup>6</sup>	410	1500°R
<u></u>			
BALANCE UTILIZED:			OOFEEICIENT
	CAPACITY:	ACCURACY:	TOLERANCE:
NF			
SF			
AF			
PN ÓL	A		
Y	A		
COMMENTS: 75	iron-constantan T/C	l's on ORB	
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TABLE I.

:

TEST RUN NUMBERS 202 IDVAR (2) K51 IDVAR (1) DATE : JUNE, THERMO COUPLE HODKUP PEREL 4~X10"6 DATA SET/RUN NUMBER COLLATION SUMMARY MODEL Ń 22 23 30 5 61 1 1 С О 60 1 ~ 1 ſ 9 COEFFICIENT SCHEDULES ļ 33 37 20 メイ 9 31 32 38 13 9 べ 17 à で 3 2 す 9 T-TANK THERMOZONOLE HOOKUP (CONST. SET 200) TABLE II. NO. OF RUNS N SCHD. PARAMETERS/VALUES Re\* Ma 5.3 52 5,3 5.2 1.0 4.0 4.0 0. 30 S 0 Q 0 0,8,8,8,3 R 2 30 8 8 8 8 3 8 9 90 0 EST : *IH 28 (AR*C 3)5-195) CONFIGURATION で ア ଷ TYPE OF DATA DATA SET 15 05 07 06 13 14 16 LT 63 90 9 20 C4 8 REVOO1 = ユ

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TEST RUN NUMBERS 202 IDVAR (2) DATE : JUNE, 1974 IDVAR (1) THERMOCOUPLE HOOKUP X ~ XIO<sup>-6</sup>PERFT 4th character of the dataset identifier describes the T/C location. wing lower surface wing upper surface DATA SET/RUN NUMBER COLLATION SUMMARY MODEL (Concluded) 36 26 33 25 ò ह 200 27 3 g OEFFICIENT SCHEDULES て 39 1 I I ſ OL - ORBITER THERHOCONDLE HOOKUP COUST SET 100 Li~TANK THERHOCOUPLE HONKUP (CONST.SETADO) 1 PARAMETERS/VALUES NO. Ref Ma | RUNS Ċ ш -TABLE II. OMS pcds chine Po 0 140 5.3 a B Ret Ma 1.052 I ပ ρ SCHD. 0 underside fuselage 0 30 2 2 2 3 9 3 20 external tank TEST : *I*H28(ARC 3.5-195) CONFIGURATION エ 0 I The ыч A TYPE OF DATA DENTIFIER DATA SET 25 23 36 27 REVOI8 61-20 2 ส 24 •

vertical tail

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canopy

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body sidewall

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TABLE III MODEL DIMENSIONAL DATA

MODEL COMPONENT: BODY - B22 GENERAL DESCRIPTION: Fuselago. Configuration 3A per Rockwell Lines VL70-000147B. NOTE: Identical to B19, except underside. Model Scale = 0.006 DRAWING NUMBER: V1.70-000147B **DIMENSIONS:** FULL-SCALE MODEL SCALE Length - in 1290.3 7.742 Max. Width - in 1.606 267.6 Max. Depth - in 1.467 244.5 Fineness Ratio 4.84601 4.84601 Area' - Ft<sup>2</sup> Max. Cross-Sectiona? 0.0139 386.67 Planform Wetted Base

TABLE III (Continued) MODEL DIMENSIONAL DATA

MODEL	COMPONENT:	Canopy	- C7

GENERAL DESCRIPTION: Configuration 3 per Rockwell Lines VL70-000139

Model Scale = 0.006

DRAWING NUMBER	<u>VL70-000139</u>		
DIMENSION:	•	FULL SCALE	MODEL SCALE
Length ( $X_0 = J$	433 to X <sub>0</sub> = 670) - in. FS	237	1.422
Max Width Mox Depth (2	Z <sub>o</sub> = to Z <sub>o</sub> = 501) - in	FS	
Fineness Ratio		-	
Area			
Max Cross	-Sectional		
Planform	•		
Wetted			
Base			، میں میں میں اور میں میں میں میں میں میں اور
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	•·· ·	•	

TABLE III (Continued) MODEL DIMENSIONAL DATA

MODEL COMPONENT:F	5 Body Flap	والإيادة فالالتجاري كإذال المتعار فتعارفه المتعاد	·····
GENERAL DESCRIPTION:	3 Configuration per R.	ockwell Lines	VL70-000139
Scale Model = 0.006			
DRAWING NUMBER	VL70-000139		•
DIMENSION:	<u>.</u>	ULL SCALE	MODEL SCALE
Length - in Max Width - in		<u>84.70</u> 267.6	.508 1.606
Mox Depth			
Fineness Ratio Area – Ft <sup>2</sup>		<u> </u>	
Mox Cross-Sec Planform	tional	142.5	.005
Wetted	·	38.0958	.0014
<b>LU3</b> 5	•		• • •

TABLE III (Continued) MCDEL DIMENSIONAL DATA

MODEL COMPONENT: ONS Pod - M4

GENERAL DESCRIPTION: Con

Configuration 3 per Rockwell Lines VL70-000139

NOTE: M4 identical to M3, except intersection to fuselage. Model Scale = 0.006 DRAWING NUMBER VL70-000139 DIMENSION: FULL SCALE MODEL SCALE 346.0 2.076 Length - IN Max Width - IN 108.0 .648 113.0 Max Depth - IN .678 Fineness Ratio Area - FT<sup>2</sup> Max Cross-Sectional Planform Wetted Base

### TABLE III (Continued) MODEL DIMENSIONAL DATA

MODEL COMPONENT: T8 - EXTERNAL TANK	· · · · · · · · · · · · · · · · · · ·	
GENERAL DESCRIPTION: 2A Configuration p VI.73-000018 and VL72-000061"C" Body of	er Rockwell Lines; Revolution	
Scale Model = 0.006		•••••••••••••••••••••••••••••••••••••••
DRAWING NUMBER VL'13-0000	18	
DIMENSION:	FULL SCALE	MODEL SCALE
Length - In. (Nose @ $X_{T} = 309$ )	186.50	1.119
Max Width (Dia) - In.	324.0	1.944
Max Depth		<del>مىرىتى بىن مىرىتى ب</del>
Fineness Ratio L/D	<u> </u>	6.1389
Area - Ft. <sup>2</sup>		
Max Cross-Sectional	572.56	0.02061
Planform		
Wetted		····
Base		
WP of tank centerline, $(Z_{\hat{T}})$ In.	400.0	2.400

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### TABLE III (Continued) MODEL DIMENSIONAL DATA

MODEL COMPONENT: VERTICAL - V 7

# GENERAL DESCRIPTION: Centerline Vertical Tail, Doublewedge Airfoil

with Rounded Leading Edge

NOTE: Same as V5, but with manipulator housing removed.

Model Scale = 0.006

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DRAWING NUMBER:	VL70-000139		
DIMENSIONS:		FULL-SCALE	MODEL SCALE
TOTAL DATA		• \	
Area (Theo) Ft <sup>2</sup> Planform		425.92	0.0153
Span (Theo) In Aspect Ratio		315.72	1.894
Rate of Taper Taper Ratio Sweep Back Angles degrees		0.507 0.404	0.507
Leading Edge Trailing Edge		<u>    45.000                              </u>	<u> </u>
Chords: Root (Theo) WP			1.611
Tip (Theo) WP MAC	•	108.47	0.651
W. P. of .25 MAC B. L. of .25 MAC	•	<u>1463.50</u> <u>635.522</u>	<u>8,781</u> <u>3,813</u> 0,00
Airfoil Section Leading Wedge Angle [	)eg	10.000	10.000
Trailing Wedge Angle Leading Edge Radius	Deg	14.920	14.920
Void Area Blanketed Area		<u>    13.17</u> <u>    0.00</u>	0.0005

### TABLE III (Concluded) MODEL DIMENSIONAL DATA

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MODEL COMPONENT: WING-W 111		
GENERAL DESCRIPTION: Configuration 3A per Rockwell	Lines VL70-0001	47B.
NOTE: Identical to W107, except lowered 3.5" and	increased cuff i	ncidence.
Model Scale = 0.006		
TEST NO.	DWG. NO. VL7	0-000147B
DIMENSIONS:	FULL-SCALE	MODEL SCALE
TOTAL DATA		•
Area (Theo.) Ft <sup>2</sup>		0.00(0
Planform	2690.00	0.0968
Span (Theo In.	<u>936.68</u>	- 5.620
Aspect Katio	1 1 77	1,177
Kaue of Taper Tapon Datio	0.200	0.200
Dihedral Angle, degrees (@ T.E. of Elevon)	3.500	3.500
Incidence Angle. degrees	0.500	0.500
Aerodynamic Twist, degraes	+3.000	+3.000
Sweep Back Angles, degrees		. *
Leading Edge	45.000	45.000
Trailing Edge	10.24	-10.24
0.25 Element Line	<u>35.209_</u> .	_35.209
Chords:	600.01	1.135
KOOT (INEO) D.P.O.V.	137.85	0.827
	174.81	2. 49
Fus. Sta. of .25 MAC	1136.89	6.821
W.P. of .25 MAC	295.70	1.774
B.L. of .25 MAC	182.13	1.093
EXPOSED DATA		
Area (Theo) Ft	1752.29	0.063
Span. (Theo) In. BP108	720.68	4.324
Aspect Ratio	2.058	2.058
Taper Ratio	0.2451	0.2451
Chords		2 371
Root BP108		0 827
$11p 1.00 \frac{p}{2}$	37.85	
MAC	393.03	2.358
Fus. Sta. of .25 MAC	<u>1185.31</u>	<u>-7,112</u>
W.P. OF .25 MAC		1 511
B.L. OT .25 MAG		
AITTOIL SECTION (NOCKNEIL AND ANDA)		
Root $\underline{b} = @ Y_0 199 to NACA 0010$	0.10	0.10
	0.12	0.12
$\frac{1100}{2}$		
Data for (1) of (2) Sides		
Leading Edge Cuff.2		

Planform Area Ft<sup>2</sup> Leading Edge Intersects Fus M. L. Ø Sta Leading Edge Intersects Wing Ø Sta C.0043 3.000 6.501

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			1.0														
	T/C	Skin		LO	CATIO	ON			T/C	Ek The	cin	<u>, /</u>		LOC	TION	h/2	2
r.	No.	Thick!	x/1	x/ c.	<u>y</u>	-+	0/2	Z	No.		2E	X/ 1 60		NC	<u> </u>		
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	2		.050			-+			36	.0	34	.70	<u>`</u> +				
	3	4	.075							.0	32	.70	2+				
	4	.034	.10			4			38	.0	35	.70					
	5	.033	.125						39	<mark>┼┈╀╴</mark> ┽		.8	25				
	6	.150							40			.8	25		OMS P	ods	
	7	.034	.034 .175 [		NDER	s	DE	ļ	41	<b> </b>		.8	25				
	8	*	.20	F	USEL	,AQ	ΕŹ	ļ	42	ļ	_	.9	<u>0</u>				
	9	.035	.25		BP =	= d	.0	ļ	43	<b> </b>	<u> </u> .	.9					
	10	*	.30					<b></b>	44		1	.9	0				
	11	.034	.40					<u> </u>	45	•	)35	1.1	0		<u> </u>		ļ
	12	.035	.50						46		4-	1.1	5		CHINE		<b>İ</b>
	13 .		.65						47			.2	0		1		·
	14	1	80						48		1	<b>.</b> 1	7		CANOF	Y	<u> </u>
	15	.036	036 .95			1		49	·	1		.425		MID FU	ISE	<u> </u>	
	16	.030	030.35		Å			50	<u> </u> .	031	11	•	.05	 	40%	<u>.</u>	
	17	.ò27	.40		UNDERSIDE				51		030			.10		40%	ö 
	18	<u>†-</u>	.50	1	FUSE	USELAGE			52	•	030			.20	ļ	409	6
	19 *		.60		BP	=	117.0	o l	53	3.029				.30		409	6
	20	17	.70	1	1	1			54	•	02 <b>8</b>	WING		.40		409	6
	21 *	.028	3 :80	1 .	1	Τ		1	55			LWI	<u> </u>	. 50		409	8
	22	.031	.90	1	-	T	1		56	*				.60		40	70
	22 *	.030	5 1.00	1		1	1		57		4			.70	 	40	76
	23	.034	4 . 30	1		Ţ	-		58	*	.029			.80		40	76
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		.03	4 .50			╉	-+		66	;#	-+-		<b>†</b>	.70		60	)%
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	34	.03	.60			1			61	3	1			.90		0	J:/0

### TABLE IV ORBITER THERMOCOUPLE LOCATIONS

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r/c T	Skin		75			VTION	b/2	2		/U 0.	Thi	ck	x/1	X	/c	y	1	b/2	Z	
No.	Thick	<u>, x</u>	1	x/c			90%	+	+-		1									
60	.034	4	1	.20			0.002	╺┼╍╍╸	-1-		1									
70		WI	ING	.40			80%	+-	+		+									
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72	.03	5	<b>I</b>	.80	2-1-		A007	+			-†									
73	.03	5	9	.20	2		40%	+-	-+-		+									
74	.03	4 W	ING	.40	2		40%							·						
75	.03	5 U	P	.60	<u>-</u>		40%	-+-			+		1					•	 	
76	.03	7	<u> </u>	.8	0		40%		-+		+-									
77	.03	4	1	1.2	0		0%				+-		+	-		Τ				
78	.03	50 W	ING	.4	0		00%				-+-		┥┷━			1				
79	.03	30		.6	0		60%	-+-			-+-		+	†		1				
80	.0	31	4_	8.	0		60%	<u>-</u>	<b></b> }		-+-		+	+						
81	.0	27	4	1.2	20		80%				-+-		+			+		1		
82	.0	28	WING		10		80%	<u>}_</u>	+				+			-				
83	.0	28	UP	<u> </u>	50	<b> </b>	80%	-+-			-+-							1	1	
84	.0	28	*		BO	ļ	80%	ő										1		
85		39	4		25	ļ		<u></u> }	.57						<u></u>	-+-		1	1	
80	1.0	40	VER	rl.	50	ļ		Þ	.57	•										
87	1.0	37	Τ		75			Þ	.57				-+			<u> </u>		+		
88		)33	VER	r .	35				.42							-+				
89		)34	*	Π.	6				.42	<b> </b>								+	-†-	
<u> </u>										<b> </b>						-+		+	-+-	
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						T									<b></b>		<u></u>		+	
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# THE OPATTER THERMOCOUPLE LOCATIONS (Concluded)

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REPRODUCIBILITY OF THE ORIGINAL PAGE D POOR.

TABLE V

EXTERNAL TANK THERMOCOUPLE LOCATIONS

T/(	SKIN	LOCA	TION <b>DEG</b>		SKIN THICK	LOCI.TI X/1		T/C NO.	SKIN THICK.	LOCAT X/1	$\phi$ DEG.	
NO 1 2 3 4 5 6 7 8 9 10 11 2 13 14 5 16 17 8 9 20 12 2 2 2 2 2 2 3 3 3 3 3 3 3 3 3 3 3 3	THTCK. 0.03 0.03 0.03 0.03 0.03 0.03 0.03 0.03 0.02 0.03 * 0.03 * 0.03	X/1 0 0.005 0.020 0.025 0.125 0.20 0.255 0.225 0.225 0.225 0.225 0.30 0.325 0.355 0.355 0.355 0.355 0.325 0.355 0.355 0.325 0.355 0.355 0.325 0.355 0.355 0.40 0.355 0.355 0.355 0.325 0.355 0.355 0.355 0.355 0.360 0.325 0.355 0.355 0.355 0.355 0.355 0.355 0.355 0.355 0.355 0.360 0.355 0.35	P DEG.      NOSE     180     180     90     12.5     90     12.5     90     12.5     90     12.5     90     12.5     90     112.5     90     12.5     90     135     12.5     90     67.5     180     135     12.5     90     135     12.5     90     135     12.5     90     135     12.5     90     135     130     135     180     157.5	10. 34 35 6 7 8 90 1 2 3 4 5 6 7 8 90 1 2 3 5 5 5 5 5 5 5 5 5 5 5 6 6 6 6 6 6 6 6	0.032     0.033     0.033     0.033     0.033     0.033     0.031     0.032     0.032     0.033     0.032     0.032     0.031     0.031     0.031     0.032     0.033     0.031     0.032     0.031     0.032     0.031     0.032     0.031	.40     0.45     0.45     0.45     0.50     0.50     0.55     0.55     0.55     0.55     0.55     0.55     0.55     0.55     0.55     0.55     0.560     0.60     0.60     0.60     0.60     0.55     0.55 <t< td=""><td>135     135     12.5     90     67.5     45     0     180     157.5     12.5     90     180     157.5     12.5     90     180     157.5     12.5     90     180     157.5     135     12.5     90     180     157.5     135     12.5     90     180     157.5     135     12.5     90     180     157.5     135     12.5     90     180     157.5     135     135     135     135     135     136     157.5     90     67.5     90     135     135  &lt;</td><td>67 68 97 70 72 73 74 * 76 77 78 99 81 82 83 84 85 86 87 88 90 91 92 93 94 95 * 99 99 99 99 99 99 99 100 1 102 103 105 105 105 105 105 105 105 105 105 105</td><td><math display="block">\begin{array}{c} 0.030\\ 0.030\\ 0.033\\ 0.033\\ 0.033\\ 0.032\\ 0.031\\ 0.030\\ 0.030\\ 0.030\\ 0.030\\ 0.030\\ 0.032\\ 0.031\\ 0.030\\ 0.031\\ 0.032\\ 0.031\\ 0.032\\ 0.031\\ 0.032\\ 0.033\\ 0.032\\ 0.032\\ 0.033\\ 0.032\\ 0.032\\ 0.033\\ 0.032\\ 0.</math></td><td>0.60 0.60 0.625 0.65 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.70 0.70 0.70 0.770 0.775 0.755 0.755 0.755 0.880 0.880 0.885 0.855 0.855 0.855 0.90 0.90 0.90 0.90 0.90 0.90</td><td><math display="block">\begin{array}{c} 45 \\ 0 \\ 180 \\ 180 \\ 157.5 \\ 135 \\ 112.5 \\ 90 \\ 67.5 \\ 135 \\ 135 \\ 157.5 \\ 135 \\ 12.5 \\ 90 \\ 67.5 \\ 135 \\ 12.5 \\ 90 \\ 67.5 \\ 135 \\ 112.5 \\ 90 \\ 67.5 \\ 135 \\ 112.5 \\ 90 \\ 67.5 \\ 135 \\ 112.5 \\ 90 \\ 157.5 \\ 135 \\ 135 \\ 112.5 \\ 90 \\ 157.5 \\ 13</math></td></t<>	135     135     12.5     90     67.5     45     0     180     157.5     12.5     90     180     157.5     12.5     90     180     157.5     12.5     90     180     157.5     135     12.5     90     180     157.5     135     12.5     90     180     157.5     135     12.5     90     180     157.5     135     12.5     90     180     157.5     135     135     135     135     135     136     157.5     90     67.5     90     135     135  <	67 68 97 70 72 73 74 * 76 77 78 99 81 82 83 84 85 86 87 88 90 91 92 93 94 95 * 99 99 99 99 99 99 99 100 1 102 103 105 105 105 105 105 105 105 105 105 105	$\begin{array}{c} 0.030\\ 0.030\\ 0.033\\ 0.033\\ 0.033\\ 0.032\\ 0.031\\ 0.030\\ 0.030\\ 0.030\\ 0.030\\ 0.030\\ 0.032\\ 0.031\\ 0.030\\ 0.031\\ 0.032\\ 0.031\\ 0.032\\ 0.031\\ 0.032\\ 0.033\\ 0.032\\ 0.032\\ 0.033\\ 0.032\\ 0.032\\ 0.033\\ 0.032\\ 0.$	0.60 0.60 0.625 0.65 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.655 0.70 0.70 0.70 0.770 0.775 0.755 0.755 0.755 0.880 0.880 0.885 0.855 0.855 0.855 0.90 0.90 0.90 0.90 0.90 0.90	$\begin{array}{c} 45 \\ 0 \\ 180 \\ 180 \\ 157.5 \\ 135 \\ 112.5 \\ 90 \\ 67.5 \\ 135 \\ 135 \\ 157.5 \\ 135 \\ 12.5 \\ 90 \\ 67.5 \\ 135 \\ 12.5 \\ 90 \\ 67.5 \\ 135 \\ 112.5 \\ 90 \\ 67.5 \\ 135 \\ 112.5 \\ 90 \\ 67.5 \\ 135 \\ 112.5 \\ 90 \\ 157.5 \\ 135 \\ 135 \\ 112.5 \\ 90 \\ 157.5 \\ 13$	
*	Data we:	re not c	btained at	these	<b>T/C</b> 1	ocations	•	108 109 110 111	0.030 0.029 0.033 0.033	0.90 0.90 0.935 0.974	67.5 45 180 180	

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	Re <sub>∞</sub> /ft × 10-6	1:0				-	0.4	1.0			0.4	0.4	0°4	1.0											-		-				
	M	5.22				<b>***</b>	5	5.22			5.3		-	5.22													-		/Cs	đix	
	Const Set**	81	200	200	200. 200.	100	100	201	100	200	200	8	200	200	100	100	500	500	100	100	200	2002	100		002	001	100	37' 28"	nel Tank T	the Appen	
	8	0						-	ŝ	ĥ	0										ندا دیر رو ا				-		-	ی ۲	Exter	nt new	
	<b>a</b> strut. deg.	0	0	-10.0	20.0	20.0	20.0	-10.0	<b>ķ</b> : 1	*	-10.0	-10.0	20.0	-20.0	-20.0	10.0	10.0	-10.0	-10.0	20.0	20.0	20.0	20.0	-10.0	-10.0	-10.0	20.0	$\alpha_{strut}$	čs: 200 -	are of	10 01 0 01
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TABLE VI RUN NUMBER/TUNNEL CONDITION SUMMARY
**TABLE VI** RUN NUMBER/TUNNEL, CONDITION SUMMARY (Concluded)

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\*\* 100 - Orbiter T/Cs; 200 - External Tank T/Cs
\*\*\* Actual test values are given in the Appendix



Figure 1. Orbiter/External Tank General Layout

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a. 50-0 Orbiter -- 147-B Configuration Thermocourle Locations Figure 2. Model Instrumentation Sketches (Concluded).

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a. Orbiter/Tank at 0.0 degrees Figure 3. Model Installation Photographs

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b. Orbiter/Tank at -60.0 degrees Figure 3. Model Installation Photographs

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c. Orbiter/Tank at 90.0 degrees Figure 3. Model Installation Photographs



d. Orbiter/Tank at 120.0 degrees Figure 3. Model Installation Photographs

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## DATA FIGURES

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8 ŀ ົດ PARAMETRIC VALUES 30.000 BETA 1.000 ::: . . œ (REVB20) 5 ALPHA LONGITUDINAL LOCATION. X.L ..... ::: ::: i;.:1 ORBITER BODY SIDEWALL, ORBITER ALONE BODY. SIDEWALL ::**:**. • . : : : • . . • : e. AMES 3.5-195 IH28 01 . MACH 5.219 2 ÷ 2 425.000 ., ТНХИАН 2850 2000.1 1.0000 FIG. 10 .1000 0100 .0010 .0001 ¥ SOD♦

RATIO OF LOCAL TO REFERENCE HEAT TRANSFER COEFFICIENTS, H/HREF

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Section 2

(REVB20) ALPHA RN/L BODY SIDEWALL AMES 3.5-195 1H28 01 MACH 5,219



2 ר דר דר ສ PARAMETRIC VALLES 60.000 BETA: 1.000 BAGE œ (REVB21) ALPHA r. ŒК LONGITUDINAL LOCATION. X/L FIG. 10 ORBITER BODY SIDEWALL, ORBITER ALONE ; BODY SIDEWALL m. AMES 3.5-195 1H28 01 MACH 5.220 Ņ 2 375.000 ---4 ----HAW/H7 -850 -850 -850 1.000 1.0000 .1000 -0100 00100 .1000. រ័ត្តលាទ RATIO OF LOCAL TO REFLRENCE HEAT TRANSFER COFFFICIENTS. HAHREF

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8 460 ດ PARAMETRIC VALUES -90.000 BETA 1.000 PAGE 8 ٠: (REVB25) 5 ALPHA . . LONGITUDINAL LOCATION. X/L . : ORBITER BODY SIDEWALL, ORBITER ALONE BODY SIDEWALL : : · • 供 .: e. đr AMES 3.5-195 1H28 01 MACH 5.219 2 z 425.co0 FIG. 10 тнумн 1938. 1932. I 0100 .1000. 000 8.... 1.0000 <sup>™</sup>O□◊

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CRBITER BODY SIJEWALL, ORBITER ALONE

FIG. 10



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g 467 ຸດ PARAMETRIC VALIÉS -30.000 BETA 1.000 PAGE • • . . ÷ : : ထ 1 : . : . : • : :: . • • • (REVB27) 1 :! 17 ALPHA-RN/L .:. ÷. : • : • LONGITUDINAL LOCATION. X/L ; ; ..... ÷. : : : FIG. 10 ORBITER BODY SIDEWALL. ORBITER ALONE ::  $\pm 1$ BODY SIDEWALL ::.: ..... ÷÷ : : 1 •1 1 : : : : . . . . . : : • • • • • ÷ : . .**m** : :: : AMES 3.5-195 IH28 01 ÷ ÷. MACH 5.220 • 1 2 • . . . . z 501.600 . НА¥/Н7 .850 .930 1.000 1.0000 .0010 .10001 -0100 1000. ¥O∏¢

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FIG. 10

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CRBITER BODY SIDEWALL, ORBITER ALONE FIG. 10

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