

COLLEGE OF ENGINEERING

Statistical Distribution of
Time to Crack Initiation and
Initial Crack Size Using Service Data

by

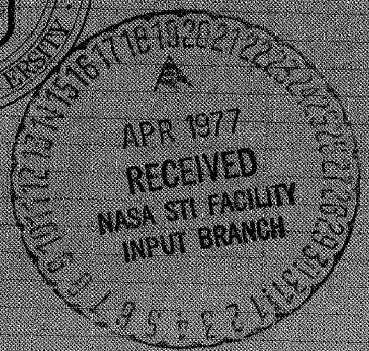
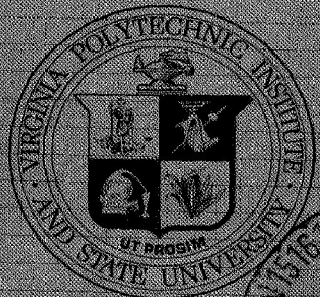
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Final Report
NASA NSG. 1099
Supplement 1.
February 1, 1977

VIRGINIA POLYTECHNIC INSTITUTE AND STATE UNIVERSITY



BLACKSBURG,
VIRGINIA

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ABSTRACT

Crack growth inspection data gathered during the service life of the C-130 "Hercules" airplane is used in conjunction with a crack propagation rule to estimate the distribution of crack initiation times and of initial crack sizes.

Because at early inspections many small cracks are missed a Bayesian statistical approach is used to calculate the fraction of undetected initiation times as a function of the inspection time and the reliability of the inspection procedure used.

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List of Symbols

a, a_i, a_0, a_u	crack length, crack initiation size = .03 in, initial crack size, ultimate crack size
A, A_k	event, observed crack length
B_k	constant
C, C^*	crack growth parameters
C_1, C_2, C_3, C_4	Constants of integration
da/dN	crack growth per cycle
da/dt	crack growth per flight hour
D, \bar{D}	detected and undetected crack
E	event
$f(1/r)$	geometric correction factor
$f_X(x), f'(x), f''(x)$	probability density function general, prior and posterior density function
$g(x)$	function
$F_X(x)$	probability distribution function
i	index
k, k_1, k_2, k_3, k_4	normalizing constants
L	geometric constant
$L(x)$	likelihood function
m_2, m_3, m_4	second, third and fourth moments of data
n, n^*	parameters of crack growth relation
$N(t)$	number of gusts per flight hr
P	probability
r	geometric constant
R	fatigue strength

List of Symbols (Contd.)

$S(t)$	laboratory flight hrs
t, t_i	inspection time, crack initiation time
$V(\cdot)$	coefficient of variation
x, y	variables
x_0	Weibull location parameter
Z_α	α th percentile for normal distribution
$\alpha, \bar{\alpha}$	Weibull shape parameters
$\beta, \bar{\beta}$	Weibull scale parameters
β_1, β_2	moment parameters
γ	Johnson S_B distribution location parameter
$\Gamma(\cdot)$	Gamma function
δ	Dirac function
Δk	stress intensity range
$\Delta\sigma$	stress range
ϵ	Johnson S_B distribution shape parameter
η	Johnson S_B distribution shape parameter
λ	Johnson S_B distribution shape parameter, Poisson failure rate
ξ	severity index

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Section I

INTRODUCTION

The Lockheed C-130 aircraft has been in use by the U.S. Air Force for more than two decades in various theaters of operation under several commands.

The center wing box of the airplane has been periodically inspected and the size of cracks growing around rivet holes in the skin has been recorded. Small cracks were permitted to grow through several inspections. Eventually cracks were repaired or in some cases the whole center wing box was replaced.

The aim of this paper is to reduce the C-130 service-flight inspection data by means of fracture mechanics and statistical analysis into a form from which the probability of time to crack initiation and the distribution of initial flaw sizes for various locations on the aircraft can be determined.

The importance of the time to crack initiation for the purposes of reliability analysis and maintenance schedules for aircraft has been indicated by numerous authors [1-8]. Though such data has been available for laboratory test specimens [9-10] they have not been computed from actual service inspection records for a large number of full scale structures. For the purposes of crack propagation analysis the time to crack initiation, defined here as the time required for a crack to grow to an inspectable size, is not needed if the initial crack size is known [11-15].

Because initial crack sizes are noninspectable they are computed from crack propagation relations by backward extrapolation [11,12].

Previous attempts to define the distribution of time to crack initiation for the C-130 and C-141 aircraft [16-18] have not yielded satisfactory results. Impellizzeri [19] discussed methods to evaluate time to crack initiation for the F-4 aircraft but the lack of sufficient data made the problem difficult. In these reports the data was not normalized, that is, the time at first crack sighting was used as the time of crack initiation regardless of crack size. This type of approach is reasonable only if inspection intervals are short so that all the cracks are small and of approximately the same size at first sighting. It is, however, impractical and uneconomical to perform inspections at such short service intervals [5,6].

The inspection records do not contain cracks smaller than .03 inches (.0762 cm) in length. Therefore, this size has been arbitrarily chosen as the minimum inspection size (crack initiation size) (a_i).

Since the objectives of this investigation are to determine the distributions of the times of growth to minimum inspection size (crack initiation) and of initial flaw sizes, crack sizes are normalized with the use of fracture mechanics. A growing crack is extrapolated backwards from its first recorded crack size to .03 inches (minimum inspection size) yielding the time to crack initiation (t_i) [6]. In a similar fashion, the crack is grown backwards to a time equal to zero to yield an initial flaw size (a_0).

All of the C-130 crack inspection data examined refer to rivet holes in the center wing box section of the aircraft (Fig. 1). The center wing box was divided into nearly one hundred inspection locations which were symmetric about the center line of the aircraft.

The original inspection data was received from Warner Robbins Air Logistic Center (ALC/ACDCJ), on a nine track magnetic tape.

The following information is available for each aircraft:

- aircraft series
- aircraft serial number
- total flying hours (at time of inspection)
- date of inspection
- military command
- military base
- facility where inspected
- number of inspection
- crack locations
- crack sizes
- crack numbers (there may be more than one crack at a location)

In order to perform the discussed computations, crack growth constants were computed for each individual location and were analyzed statistically.

Two approaches were used to describe the rate of crack growth at various locations. One is the more common practice of fixing the crack growth parameters. The second approach is to let the crack growth parameters vary from location to location even though the material, geometry and type of loading are similar between locations.

Only cracks that grew were utilized in the present analysis. These comprise approximately fifteen percent of the total number of rivet holes and consequently conclusions reached are valid for this reduced population.

Statistical distributions were fitted to time to crack initiation and to initial flaw sizes in order to describe their expected behavior at each location. Since only the cracked population was considered, probabilities are conditional on the presence of a crack [20].

The influence of the modes of operation within the various commands is reflected in the rates of crack propagation indicating more or less severe usage. The data has, as a consequence, been also used to estimate the parameters of the load distributions applied to the airplanes.

Because the method of inspection is not a hundred percent reliable some cracks present in the structure were presumably not discovered. Utilizing an assumed inspection reliability curve combined with a Bayes' approach, the distribution of crack initiation times undetected during an inspection has been estimated [21,22].

Section II
METHODS OF ANALYSIS

1. Time to Crack Initiation and Initial Flaw Size

Inspection of the aircraft took place at irregular intervals. As a result raw inspection data points could not be directly compared because the length of time during which cracks have propagated differed for each crack.

Once the crack propagation parameters have been calculated, cracks could be mathematically grown to either a common size or to a common inspection time. Such a "normalization" procedure permits the determination of statistical distributions for "crack initiation times" defined here as the time at which the crack size was .03 in (minimum inspectable crack length), and for "initial crack sizes", the size of cracks at time equal to zero (Fig. 2).

In order to determine the time to crack initiation (t_i) and the initial flaw size (a_0) the "Power Law" crack growth relation [23]

$$da/dN = C(\Delta k)^n \quad (1)$$

was used.

$$\text{Here } \Delta k = \Delta\sigma\sqrt{\pi a} f(1/r) = \text{Stress Intensity Range} \quad (2)$$

da/dN = crack growth per cycle

C & n = material constants

$\Delta\sigma$ = stress range

a = crack length

$f(1/r)$ = geometric correction factor

Eq. 1 may be rewritten utilizing Eq. 2 and a rate-of-cycling term, dN/dt as

$$da/dt = C \left(\Delta\sigma\sqrt{\pi a} f(1/r) \right)^n dN/dt \quad (3)$$

$$= C \left(\Delta\sigma\sqrt{\pi} f(1/r) \right)^n dN/dt a^{n/2}$$

$$\text{Letting } C^* = C \left(\Delta\sigma\sqrt{\pi} f(1/r) \right)^n dN/dt \quad (4)$$

and

$$n^* = n/2 \quad (5)$$

where C^* is in units of [$\text{in.}/\text{in.}^{n/2}/\text{flight hours}$], Eq. 3 becomes

$$da/dt = C^* a^{n^*} \quad (6)$$

Eq. 6 is the crack growth relation used in this analysis. Taking the logarithm of Eq. 6 a straight line relation, whose slope is n^* and intercept is $\log C^*$, is obtained.

$$\log (da/dt) = \log C^* + n^* \log a \quad (7)$$

Several methods were tried in attempts to evaluate the parameters of Eq. 6. To determine the crack growth constants C^* and n^* for each growing crack individually the average crack growth rate between two inspections is calculated as the difference in crack sizes divided by the inspection interval evaluated at the average of the two crack sizes and at the mid point of the interval. Because two constants are to be determined a minimum of three successive inspections during which a crack grew are required to produce two points for Eq. 7 (Figs. 2 & 3). Not many cracks were permitted to grow through three inspections.

Due to this limitation only a relatively small number of the voluminous data could be utilized to obtain crack growth parameters for individual cracks.

2. Linear Least Squares Regression

In order to increase the number of available data points an attempt was made to evaluate crack growth rate versus crack length data for each pair of successive inspections and then to use all of these values at a specific location on the aircraft for individual commands in a linear regression analysis as shown in Fig. 4.

In a first attempt all the da/dt vs. a_{mean} data were utilized in order to find characteristic n^* and C^* values for each location. A computer program (Statistical Analysis System) was used to perform a linear least squares fit to Eq. 7. The slope and intercept of the line are the n^* and C^* values, respectively.

It was found that the exponent, n^* , was lower than what had been anticipated from the examination of aircraft aluminum crack growth curves [11] and varied considerably. An n^* of approximately 1.5 ($n=3$) had been expected. It is recognized that n^* should be nearly constant in value since similar material and geometries are used in each aircraft and location. Referring to Eq. 4, C^* varies with material constant C , material constant n , geometric correction factor $f(1/r)$ and stress range $\Delta\sigma$.

The variations in n^* may be explained as follows: Fig. 5 represents two ideally growing cracks in the same location but on two different aircraft within the same command. The only difference between the two is

that aircraft B has experienced higher stress ranges than aircraft A. Note that both lines have approximately the same slope, n^* , but differ in C^* (Table 1) reflecting the difference in stress range. On the other hand a linear least squares fit of all data points (six) from both aircraft results in an $n^* = .92$ instead of an average value of 1.01. If each crack is treated individually and average values of n^* and C^* are calculated less error is introduced.

3. Non-linear Regression

In another attempt to use all available data directly the rate of growth equation (Eq. 6) was integrated

$$da/dt = C^*a^{n^*} \quad (6)$$

therefore $a^{-n^*} da = C^* dt$

Integrating both sides

$$\int_{a_i}^a a^{-n^*} da = C^* \int_{t_i}^t dt$$

$$\frac{1}{1-n^*} \left(a^{(1-n^*)} - a_i^{(1-n^*)} \right) = C^* (t - t_i)$$

$$a^{(1-n^*)} = C^*(1-n^*) (t - t_i) + a_i^{(1-n^*)}$$

$$a = \left[C^*(1-n^*) (t - t_i) + a_i^{(1-n^*)} \right]^{\frac{1}{1-n^*}} \quad (8)$$

where t_i = time to crack initiation

a_i = crack length at crack initiation = .03 in. is obtained (Fig. 6).

By assuming certain values of n^* Eq. 8 may be simplified:

for $n^* = 2$

$$\frac{1}{a} = -C^* (t-t_i) + \frac{1}{a_i}$$

$$\frac{1}{a} = C_1 t + C_2 \quad (9)$$

where $C_1 = -C^*$

$$C_2 = C^* t_i + \frac{1}{a_i}$$

For $n^* = 1.5$

$$\frac{1}{a} = \left[-\frac{C}{2} (t-t_0) + \frac{1}{\sqrt{a_0}} \right]^2$$

carrying out the operations and factoring yields

$$\frac{1}{a} = C_1 t^2 + C_2 t + C_3 \quad (10)$$

where

$$C_1 = \frac{C^{*2}}{4}; \quad C_2 = \frac{C^*}{\sqrt{a_i}} - \frac{C^{*2}}{2} t_i$$

$$C_3 = \frac{C^{*2}}{4} t_i^2 - \frac{C^* t_i}{\sqrt{a_i}} + \frac{1}{a_i}$$

Equations 9 and 10 were evaluated again using regression analysis for all a versus t data for a given location in each command (Fig. 6). Results were again unsatisfactory because n^* values were afflicted with a large scatter.

4. Conjugate Gradients Method

The next approach was an attempt to minimize the square difference between observed and predicted crack length by the method of conjugate gradients.

The function to be minimized may be written as

$$F = \sum_{k=1}^N (A_k - a_k)^2 = \sum_{k=1}^N \left\{ A_k - \left[a_i^{(1-n^*)} + C^* (1-n^*) (t_k - t_i) \right]^{\frac{1}{1-n^*}} \right\}^2 \quad (11)$$

where A_k = observed crack length
 a_k = predicted crack length
 t_k = observed time

To minimize the function, its partial derivatives

$$\frac{\partial F}{\partial C^*} = \sum_{k=1}^N -2 (t_k - t_i) \left[A_k - B_k^{\left(\frac{1}{1-n^*}\right)} \right] B_k^{\left(\frac{n^*}{1-n^*}\right)} \quad (12)$$

$$\frac{\partial F}{\partial t_i} = \sum_{k=1}^N 2C^* \left[A_k - B_k^{\left(\frac{1}{1-n^*}\right)} \right] B_k^{\left(\frac{n^*}{1-n^*}\right)} \quad (13)$$

$$\frac{\partial F}{\partial n^*} = \sum_{k=1}^N \frac{2n^*}{(1-n^*)^2} \left[A_k - B_k^{\left(\frac{1}{1-n^*}\right)} \right] B_k^{\left(\frac{1}{1-n^*}\right)} \ln B_k \left[a_i^{(1-n^*)} \ln a_i + C^*(t_k - t_i) \right] \quad (14)$$

are set equal to zero and the nonlinear equations, Eqs. 12-14 are solved simultaneously.

In the above relations

$$B_k = a_i^{(1-n^*)} + C^*(1-n^*) (t_k - t_i)$$

The gradients, $\frac{\partial F}{\partial C^*}$, $\frac{\partial F}{\partial t_i}$ and $\frac{\partial F}{\partial n^*}$, were used in a computer program (SSP subroutine FMCG). It was not possible to set bounding values in the subroutine. Therefore, certain physical limitations, such as positive crack length, were violated. The term, $B_k^{(\frac{1}{1-n^*})}$, represents crack length and was in some cases forced into the negative region by the FMGC program in order to minimize the function. The method was consequently abandoned.

5. Variable n^* Method

Crack growth parameters for individual cracks were also calculated and were then averaged to find representative C^* and n^* values.

For this method only those da/dt versus a_{mean} data were used in which a crack grew through at least two inspections (three points on an a versus t curve). Since stress ranges could vary greatly from one inspection to another, not all n^* and C^* values calculated were acceptable. Therefore, limitations were set on n^* ($1.0 < n^* < 2.5$), and on C^* ($.05 > C^*$). Many of the cracks that grew sufficiently to be investigated yielded n^* and C^* outside the acceptable range.

The acceptable parameters for cracks within a given location and command were averaged to yield mean values of $\overline{n^*}$ and $\overline{C^*}$. These average parameters were then used to extrapolate all cracks that grew, from their first sightings backwards to the time of crack initiation (t_i) and to initial flaw size (a_0).

Table 2 lists the raw data of crack length and inspection times for the relevant locations.

This method yielded mean values of n^* from approximately 1.2 to 1.9 (Table 3). When these n^* values were used to calculate a_0 , the initial flaw size varied with n^* ; a lower $\overline{n^*}$ produced a smaller $\overline{a_0}$ (see Figs. 7 and 8). Since initial flaw size in an aircraft should be independent of the life of the aircraft, the values of $\overline{a_0}$ at various locations should be of the same order of magnitude. These results indicated that n^* should indeed be a fixed material property.

6. Constant n^* Method

A value for the material property n was selected after evaluating crack growth data on 7075 Aluminum from the Rockwell International Fracture Mechanics Data Bank [9] and comparing this to a NASA Langley study on laboratory crack growth of C-130 components [10]. Based on these studies, $n=3$, i.e., $n^*=1.5$, was chosen.

With n^* established, only one additional parameter needs to be determined, hence individual crack growth data, da/dt versus a_{mean} could be used (Fig. 9). It has been mentioned previously that when n^* and C^* were both allowed to vary, the values of n^* and C^* were frequently outside an acceptable range. By fixing n^* and letting only C^* vary, the results yielded a reasonable C^* value every time. Due to the increase in the number of reasonable C^* 's for all crack locations, more crack locations could be evaluated.

The values of C^* for the various growing cracks within a location were averaged to yield a $\overline{C^*}$. This $\overline{C^*}$ value and $n^*=1.5$ were used to extrapolate the \underline{a} versus \underline{t} data of the first crack that grew at a particular location

on an aircraft. The resulting values of t_i and a_0 are shown in Table 3. It might be noted that all the $\overline{a_0}$ values for the various locations were, as expected, of the same order of magnitude (see Section III.2).

Section III

STATISTICAL DESCRIPTION OF THE DATA

Statistical distribution functions have been fitted to the times to crack initiation and to initial flaw sizes. The probability of time to crack initiation will be discussed first.

1. Time to Crack Initiation Distributions

a. Two Parameter Weibull Distribution

Time to crack initiation (or failure) has in the past been described in terms of the Weibull distribution [24]. The two parameter Weibull probability of failure is given as

$$F_{t_i}(x) = 1 - e^{-(x/\beta)^\alpha} \quad (15)$$

where $\alpha = 1.2 (\sigma/\bar{x})$ is the shape parameter

$\beta = \bar{x} / \Gamma(1 + 1/\alpha)$ is the scale parameter

\bar{x} = mean

σ = standard deviation

This distribution fits most of the data but fails to fit the end points (see Figs. 10 through 17 and Table 4). The lower portion of the data (earliest crack initiation) is considered extremely important, therefore it is necessary to try other statistical models that might fit these points better.

b. Three Parameter Weibull Distribution

The three parameter Weibull distribution

$$F_{t_i}(x) = 1 - e^{-\left(\frac{x-x_0}{\beta-x_0}\right)^\alpha} \quad (16)$$

has also been used. Because of the possible existence of flaws at time $t=0$ this distribution function will have a negative lower limit, x_0 and Eq. 16 will have to be restricted to values for $t_i > 0$. Consequently the distribution of crack initiation times becomes:

$$F_{t_i}(x) = 0 \quad \text{for } x < 0 \quad (17a)$$

$$F_{t_i}(0) = 1 - e^{-\left(\frac{-x_0}{\beta-x_0}\right)^\alpha} \quad \text{for } x = 0 \quad (17b)$$

and

$$F_{t_i}(x) = 1 - e^{-\left(\frac{x-x_0}{\beta-x_0}\right)^\alpha} \quad \text{for } x > 0 \quad (17c)$$

that is, the probability that the crack initiation time is zero has a fixed, nonzero value. The corresponding density function consists of a Dirac function at $t_i = 0$ with an area given by Eq. 17b and a truncated density function [21]. These functions are schematically shown in Figs. 18 and 19.

Such distributions have also been fitted to the data. There is significant improvement over the fit of the two parameter function, but for very short initiation times this distribution does not exhibit a good fit either. The distributions are presented in Figs. 20-27 and Table 5.

c. Johnson Distributions

Johnson distributions were examined next, due to the fact that the Johnson S_B distribution has four adjustable parameters [24]. In

order to determine which one of three types of Johnson distributions would be applicable, values of β_1 and β_2 were calculated for each data set.

$$\beta_1 = \left(\frac{m_3}{(m_2)^{3/2}} \right)^2 \quad (18)$$

$$\beta_2 = \frac{m_4}{(m_2)^2} \quad (19)$$

where m_2 , m_3 , and m_4 are second, third and fourth moments of the data, respectively:

$$m_2 = \frac{1}{n} \sum_{i=1}^n (x_i - \bar{x})^2 \quad (20)$$

$$m_3 = \frac{1}{n} \sum_{i=1}^n (x_i - \bar{x})^3 \quad (21)$$

$$m_4 = \frac{1}{n} \sum_{i=1}^n (x_i - \bar{x})^4 \quad (22)$$

From an examination of Table 4 and Fig. 28 [24] it is noted that the Johnson S_L or S_B distributions apply.

The S_B distribution has two shape parameters, γ and η , one location parameter (lower bound), ϵ , and one scale parameter, λ . The distribution is bounded by ϵ and $\epsilon + \lambda$. In the case of time to crack initiation, the range, λ , is unknown but can be estimated by the following equation:

$$\lambda = (x_{0.5} - \epsilon) \frac{(x_{0.5-\epsilon}) (x_{\alpha-\epsilon}) + (x_{0.5-\epsilon}) (x_{1-\alpha-\epsilon}) - 2 (x_{\alpha-\epsilon}) (x_{1-\alpha-\epsilon})}{(x_{0.5-\epsilon})^2 - (x_{\alpha-\epsilon}) (x_{1-\alpha-\epsilon})} \quad (23)$$

where $x_{0.5}$ = median of the data

x_{α} = α percentile of data

$x_{1-\alpha}$ = $(1-\alpha)$ percentile of data

Since primary interest lies in fitting the extreme data points, $\alpha = 1\%$ was used in most cases. More extreme values were often necessary to assure that λ is positive. In some cases the term $(x_{0.5-\epsilon})^2 - (x_{\alpha-\epsilon})(x_{1-\alpha-\epsilon})$ becomes negative if extreme values are not chosen. In the case of location 74-90 TAC, an x_{α} value had to be extrapolated below the lowest data point to obtain a positive λ .

Once ϵ and λ are established, the two shape parameters may be calculated as follows:

$$\eta = \frac{Z_{1-\alpha} - Z_{\alpha}}{\ln \left[\frac{(x_{1-\alpha-\epsilon})(\epsilon + \lambda - x_{\alpha})}{(x_{\alpha-\epsilon})(\epsilon + \lambda - x_{1-\alpha})} \right]} \quad (24)$$

$$\gamma = Z_{1-\alpha} - \eta \ln \left(\frac{x_{1-\alpha} - \epsilon}{\epsilon + \lambda - x_{1-\alpha}} \right) \quad (25)$$

where Z_{α} = the standard normal variate for α 100th percentile

$Z_{1-\alpha}$ = the standard normal variate for $(1-\alpha)$ 100th percentile

Once ϵ , λ , η , and γ have been calculated for each set of data, the following density function can be numerically integrated to find the distribution function

$$f_{t_i}(x) = \frac{\eta}{\sqrt{2\pi}} \frac{\lambda}{(x-\epsilon)(\lambda-x+\epsilon)} \exp \left\{ -\frac{1}{2} \left[\gamma + \eta \ln \left(\frac{x-\epsilon}{\lambda-x+\epsilon} \right) \right]^2 \right\} \quad (26)$$

The results of the Johnson S_B distribution for time to crack initiation are shown in Figs. 10 to 17 and the parameters for each location are presented in Table 4.

Because x_α and $x_{1-\alpha}$ may be arbitrarily chosen this family of distribution functions can be made to fit the short initiation time data points, but at the expense of missing the large number of data points in the central region.

d. Statistical Distribution Based on Service Load and Fatigue Performance

Because none of the afore discussed distribution functions fit the data equally well in the central region and the extremes, a technique based essentially on the concept of stress-strength interference will also be examined [6].

It is assumed that the fatigue strength, R , of the structure in a laboratory test can be described by a three parameter Weibull distribution. The probability that the fatigue life is shorter than a given value, x is written as

$$F_R(x) = 1 - e^{-\left(\frac{x-x_0}{\beta-x_0}\right)^\alpha} \quad (27)$$

where R and hence x is measured in laboratory hours.

The statistical density function of the number of gusts that produce significant fatigue damage in the aircraft in t flying hours has been described by Yang [6] as a continuous version of the Poisson process

$$f_{N(t)}(x_1) = (\lambda t)^{x_1} e^{-\lambda t} / \Gamma(x_1+1) \quad 0 < x_1 < \infty \quad (28)$$

where the random variable, $N(t)$, is the number of gusts in t flight hrs,

λ is the average number of significant gusts per flight hr, λt is the average number in t flight hrs and $\Gamma(\cdot)$ is the gamma function.

Because the variance of the Poisson distribution is also λt , the coefficient of variation which is a measure of dispersion becomes

$$V_{N(t)} = 1/\sqrt{\lambda t} \quad (29)$$

The number of gusts $N(t)$ experienced by an aircraft in t flight hrs can be related to the number of equivalent laboratory hrs, $S(t)$ as

$$S(t) = N(t)/\lambda \xi \quad (30)$$

where ξ is a so-called severity index that in the present analysis is assumed to be unity. As a consequence the laboratory spectrum is equivalent to the average flight spectrum.

The density function, of gust loading, Eq. 28 can be transformed with the aid of Eq. 30 into a density function in terms of equivalent laboratory flight hours

$$f_{S(t)}(x) = \lambda t^{x\lambda} e^{-\lambda t} \frac{\lambda}{\Gamma(\lambda x + 1)} \quad (31)$$

The probability that a crack will initiate in t flight hours is equivalent to the probability that the fatigue strength, R , is exceeded by the service loading converted to laboratory flight hrs, $S(t)$

$$F_T(t) = P[R \leq S(t)] \quad (32)$$

Eq. 32 is a form of the classical stress-strength interference relation [25] and can be expressed as

$$F_T(t) = \int_0^{\infty} F_R(x) f_{S(t)}(x) dx \quad (33)$$

Where $F_R(x)$ is given by Eq. 27 and $f_{S(t)}(x)$ by Eq. 20. Therefore

$$F_T(t) = \int_0^{\infty} \left[1 - e^{-\left(\frac{x-x_0}{\beta-x_0}\right)^\alpha} \right] \frac{\lambda(\lambda t)^{\lambda x} e^{-\lambda t}}{\Gamma(\lambda x + 1)} dx \quad (34)$$

For convenience of integration a change of variables is introduced:

$$y = \lambda x \quad , \quad dy = \lambda dx \quad (35)$$

Substituting into Eq. 34 the probability of failure becomes

$$F_T(t) = \int_0^{\infty} \left[1 - e^{-\left(\frac{y-x_0\lambda}{\beta\lambda-x_0\lambda}\right)^\alpha} \right] \frac{(\lambda t)^y e^{-\lambda t}}{\Gamma(y+1)} dy \quad (36)$$

Eq. 36 is a convolution integral that requires numerical integration. The values of α , β and x_0 are estimated by fitting a three parameter Weibull distribution to the central part of the crack initiation time data (Table 5).

Because $1/\sqrt{\lambda\beta}$ is a measure of the dispersion in β hrs, a small value of λ produces a large dispersion. Choosing λ values by trial and error ($\lambda\beta$ is usually less than 100) Eq. 36 is evaluated and is

fitted to the crack initiation time data so that the curve approaches the data in the lower tail (short initiation times) of the distribution as shown in Figs. 29 to 36.

Smaller values of λ tend to bend the lower tail upward and the upper tail downward.

Because short crack initiation times are of interest, the poorer fit at the upper tail is not considered significant.

2. Initial Flaw Size Distributions

It is not unexpected that material imperfections, manufacturing methods and the limitations of inspection techniques produce a certain number of undetected minute cracks even before the aircraft is put into service. Once the average parameters of the crack growth relation are evaluated and t_i is calculated for each crack, Eq. 8 can be further extrapolated backwards to time $t=0$ to yield an initial crack size as shown in Fig. 2. Only data with extrapolated initial crack sizes greater than zero can be used.

Initial flaw sizes were plotted on normal probability paper. The plots indicated that the distribution functions were not normal; the data did not fall on straight lines. As an alternative the Johnson S_B distribution was again employed. Since .03 inches (.0762 cm) had been chosen as the crack initiation size, the nondefect population (those locations with initial flaws less than .03 inches) ranges from 0.0 to 0.03 inches. Thus, $\epsilon = 0.0$ and $\lambda = 0.03$. The shape parameters η and λ , were found as previously shown from Eqs. 24 and 25 with $\alpha = 1-2\%$.

The parameters are listed in Table 6. Distribution functions, are presented in Figs. 37 to 44. It is seen that the Johnson S_B distribution fits these extrapolated points very well. Because initial flaw sizes should be independent of loading their distributions should be similar. It is seen from Table 6 that the means and standard deviations are of similar magnitude. As a consequence all initial crack sizes could be considered to belong to the same population. An overall distribution function has been derived for the combined data and is presented in Fig. 45. The mean initial crack size is .00432 in. and has a coefficient of variation of .2%.

The two Commands considered here, TAC (Tactical Air Command and PACAF (Pacific Air Command), had 263 and 223 C-130 Aircraft inspected respectively.

Since two symmetric locations and two holes at each location were all considered to be identical the total number of potential crack sites at one location is 1052 and 892 for TAC and PACAF respectively. Of these 16% in TAC and 30% in PACAF have actually developed growing cracks at the most vulnerable location (75-91, see Table 3). Aircraft operated by PACAF have a greater proportion of growing cracks at all locations while aircraft in the Tactical Air Command developed fewer inspectable and hence extrapolatable cracks. This fact would indicate that PACAF aircraft have been subjected to more severe loading conditions. Such a conclusion is not always born out by the data shown in Table 3. The value of C^* for locations 75-91 is greater for PACAF than for TAC but at the same time other locations indicate an opposite trend.

Section IV

THE RELIABILITY OF INSPECTION

1. Distributions of Detected and Undetected Cracks

The probability that an existing crack is detected is a function of a number of circumstances. These include: a.) The method of detection such as radiography, thermography, dye penetrant, ultrasonic or visual inspection, etc., and b.) The sensitivity and accuracy of the instrument or the alertness of the inspector. These conditions are influenced primarily by the size of the crack at the time of inspection.

As a consequence not all cracks are found and the probability, $P[D|a]$, that a crack is detected, given that its length is a , is a function of the crack size.

The methods of inspection or the reliability of the procedure used for crack detection in C-130 airplanes are not available. Data can however be found in the literature on the probability of crack detection in aluminum using various ultrasonic techniques [26]. A set of data from Ref. 26 is shown in Fig. 46 with a Weibull type probability function

$$P[D|a] = 1 - e^{-\left(\frac{a}{\bar{\beta}}\right)^{\bar{\alpha}}}, \quad (37)$$

where $\bar{\alpha} = 2.0$ and $\bar{\beta} = .075$ in. (.19 cm), fitted to it. Similar probability functions have been incorporated in wing failure analyses and have been reviewed by Eggwerts [27].

The C-130 inspection data indicates that cracks shorter than .03 in have not been found. The assumed probability function used here predicts

that a small fraction, 15%, of such cracks are located, while the probability of detecting a crack .2 in. in length is almost unity.

If the distribution of crack sizes prior to inspection is known, the distributions of detected and undetected cracks may be calculated utilizing the probability of detection function, Eq. 37 in conjunction with Bayes theorem [28,29].

The probability that a flaw of size, a , will be between x and $x+dx$ given that the flaw is detected may be written as

$$P[x < a < (x+dx) | D] = \frac{P[D | x < a < (x+dx)] P[x < a < (x+dx)]}{P[D]} \quad (38)$$

where D stands for detection, $P[D | x < a < (x+dx)]$ is the probability of detecting a flaw of size a (from Eq. 37), $P[x < a < (x+dx)]$ is the probability that the crack size is a and $P[D]$ is the total probability of detection independently of size. Eq. 38 may be expressed in terms of density functions as

$$f_a[x|D] = k_1 P[D|a] f_a(x) \quad (39)$$

in which $f_a(x)$ is the density function of flaw sizes prior to detection, $f_a[x|D]$ is the density function of detected flaw sizes (posterior density), $P[D|a]$ defined earlier, is called the likelihood function in Bayes' theorem terminology and $k_1 = 1/P[D]$ is a normalizing constant that will make the area under the $f_a(x|D)$ curve equal to unity.

In a similar manner the density function of undetected cracks may also be derived:

$$f_a[x|\bar{D}] = k_2 P[\bar{D}|x] f_a(x) \quad (40)$$

where \bar{D} stands for no detection, $P[\bar{D}|x] = 1 - P[D|x]$, and k_2 is a second normalizing constant. The above technique has been utilized by Davidson [21] in a slightly different form with exponential detectability and crack size distribution functions assumed.

Adding Eqs. 39 and 40

$$\frac{f_a(x|D)}{k_1} + \frac{f_a(x|\bar{D})}{k_2} = f_a(x) \quad (41)$$

indicates that the reciprocal normalizing constants, $1/k_1$ and $1/k_2$ represent the fractions of detected and undetected cracks.

Because the density function of cracks before inspection is usually unknown, the posterior function, $f_a[x|D]$, of detected cracks may be used to evaluate the prior, $f_a(x)$, and the density of undetected cracks, $f_a[x|\bar{D}]$, from Eqs. 39 and 40 [29].

As indicated by Eq. 8 crack size is a function of the parameters a_i and t_i and the time of inspection t . As a result Eqs. 39-40 are also dependent on these quantities. Since inspections were carried out at irregular intervals the above relations could only be utilized if detected cracks were mathematically grown to a common inspection time. Alternatively Eqs. 38-41 can be transformed to yield density functions of crack initiation times, t_i , at a common crack size, $a_i = .03$ in (.76 cm).

2. Distributions of Detected and Undetected Crack Initiation Times

Eqs. 39 and 40 may be rewritten in terms of crack initiation times, t_i as

$$f_{t_i}[x|D] = k_3 P[D|t_i] f_{t_i}(x) \quad (42)$$

and

$$f_{t_i}[x|\bar{D}] = k_4(1-P[D|t_i])f_{t_i}(x) \quad (43)$$

where $f_{t_i}[x|D]$ and $f_{t_i}[x|\bar{D}]$ are the density functions of detected and undetected crack initiation times, $P[D|t_i]$ is the likelihood function for detection and $f_{t_i}(x)$ is the prior density function of all crack initiation times.

$P[D|t_i]$ is derived from Eq. 37 by substitution of Eq. 8. Hence

$$P[D|t_i] = 1 - \exp \left\{ - \left[\frac{1}{\beta^{\frac{\alpha}{1-n^*}}} \left(C^*(1-n^*)(t-t_i) + a_i^{(1-n^*)} \right)^{\frac{\alpha}{1-n^*}} \right] \right\} \quad (44)$$

It should be noted that Eq. 44 becomes equal to unity when

$$(t - t_i) = - \frac{a_i^{(1-n^*)}}{C^*(1-n^*)} \quad (45)$$

because within this time interval a crack will grow to infinite length. Hence $P[D|t_i]$ will be equal to Eq. 44 for $(t-t_i) < -[a_i^{(1-n^*)}/C^*(1-n^*)]$ and becomes equal to unity for $(t-t_i) \geq -[a_i^{(1-n^*)}/C^*(1-n^*)]$, which indicates that the probability of detecting an initiation time, t_i , depends on the time of inspection, t . As a consequence the density functions of detected and undetected initiation times will also become functions of the inspection time while the density function of all initiation times should be independent of when the inspection is carried out.

The distributions of t_i described in Section II were derived from long-term records, in some instances covering 7000 hrs of flying time.

It is therefore not unreasonable to assume that the great majority of cracks have grown to inspectable size and that distributions of crack initiation times derived from these by backward extrapolation are good approximations of the total initiation time distributions. Such an assumption is conservative because long initiation times (small cracks) make up the majority of still undetected initiations.

With this assumption it is possible to determine the density functions of discovered and undiscovered initiation times for an inspection carried out earlier in the life of the structure.

The three parameter Weibull distributions, Eqs. 17, fitted to the derived data of Section II (the convolution integral representation is mathematically more complicated)

$$f_{t_i}(x) = \left[1 - e^{-\left(\frac{x-x_0}{\beta-x_0}\right)^\alpha} \right] \delta(x) + \frac{\alpha}{\beta-x_0} \left(\frac{x-x_0}{\beta-x_0}\right)^{\alpha-1} e^{-\left(\frac{x-x_0}{\beta-x_0}\right)^\alpha} \quad (46)$$

will be used where $\delta(x)$, is a Dirac function.

For specified inspection time, t , crack growth parameters C^* and n^* and $a_i = .03$ in., Eqs. 44 and 46 substituted into Eqs. 42 and 43 yield the density functions of detected and undetected initiation times. As in the case of detected and undetected cracks the reciprocal normalizing coefficients add to unity

$$\frac{1}{k_3} + \frac{1}{k_4} = 1 \quad (47)$$

and indicate the fractions of discovered and undiscovered initiation times.

The results of such calculations are shown at location 76-92, TAC Command in Figs. 47-49 for inspection times $t = 2500, 4000, \text{ and } 5000$ hrs. The parameters of the distribution functions are presented in Table 7.

It should be noted that the density functions for detected and undetected initiation times have been divided by their respective normalizing constants so that the areas under the two curves are proportional to the fractions of discovered and undiscovered initiation times and the two areas sum up to the area of the density function for all initiation times.

An examination of Figs. 47-49 indicates an expected trend. If inspection is performed too early, the majority of initiation times will remain undetected while more and more crack initiation times are found at later inspections.

The foregoing discussion has utilized longterm inspection data to derive the prior distribution of crack initiation times. Such records are however usually unavailable for aircraft in service. On the contrary at a particular inspection time the distribution of observed crack initiation times is actually the posterior distribution, $f_{t_i}(x|D)$, while the prior density, $f_{t_i}(x)$, of all initiation times and the density function of undetected initiation times, $f_{t_i}(x|\bar{D})$, are unknown.

Because the latter two functions are of primary interest Eqs. 42 and 43 may be rewritten using Eq. 47 as

$$f_{t_i}(x) = \frac{f_{t_i}[x|D]}{k_3 P[D|t_i]} \quad (48)$$

and

$$f_{t_i}(x|\bar{D}) = \frac{(1-P[D|t_i])f_{t_i}[x|D]}{(k_3-1) P[D|t_i]} \quad (49)$$

These equations would predict the density functions of all crack initiation times as well as that of the undetected times provided that inspections are carried out at approximately the same number of flight hours on all aircraft in a fleet.

Section V
CONCLUSIONS

The methods presented indicate the feasibility of determining the statistical distributions of initial crack sizes and times to crack initiation using backward extrapolation of inspection data. It appears that of the two crack propagation parameters the value of n^* is a material constant with a value of $n^* = 1.5$ while the second parameter C^* , in addition to being dependent on material and geometry, is also a function of the stress range.

Evaluation of C^* for each individual growing crack and averaging such values for specific crack locations and military commands yields satisfactory distributions of crack initiation times and initial crack sizes.

Crack initiation times are dependent on both the initial crack size and on C^* , the latter being dependent on location and load, while the distribution of initial crack sizes is independent of these factors.

The three parameter Weibull distribution fits the crack initiation times reasonably well. The convolution integral, that involves the three parameter Weibull distribution of fatigue strength and the Poisson distribution of gust occurrences does not improve the fit appreciably because this method has been developed for the purpose of evaluating laboratory fatigue tests under simulated load spectra rather than for the analysis of service data. Two parameter Weibull or Johnson S_B distributions do not fit the data very well.

The distributions of initial crack sizes on the other hand are well represented by the Johnson S_B distribution. The methods presented permit evaluation of a single initial crack size distribution for all growing cracks independently of the different loads in the two commands or the location of cracks on the wing. Though only four locations were analyzed it is believed that the distribution of initial crack sizes is representative of other locations as well. The information can consequently be used for the prediction of crack growth for the C-130 aircraft as well as for structures made of the same material with similar geometry and having undergone similar manufacturing processes. It is recognized that the distributions of time to crack initiation and initial crack size are not independent of each other but are interrelated through Eq. 8. Consequently a probability transformation can be performed on one of the two distributions to derive the density function of the other. It is obvious that such a transformation will result in distribution functions different from those presented here though it is expected that the derived function would be well approximated by the fitted curves. A transformation will be carried out at a future date.

The Bayesian approach presented here illustrates a method for estimating the proportion of undiscovered crack initiation times at various inspections and indicates the influence of the reliability of crack detection methodology particularly during early inspections when cracks are small.

Because the information concerning the reliability of the particular method of nondestructive examination used for the C-130 aircraft was not available an assumed likelihood function was used. For future systems

the documentation of NDE reliability should become an integral part of service inspection.

A similar approach may be used to determine the fraction of undetected crack sizes at a particular inspection time. For this purpose cracks would have to be analytically grown to common inspection times and their distributions analyzed. Such work may also be carried out in the future.

The present analyses have utilized average values of the crack propagation parameter C^* . The statistical distributions of the parameter have not been evaluated. Again calculations, with C^* as a random variable, may be performed at a later date.

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TABLE 1

Crack Growth Constants for Two Hypothetical Aircraft

Aircraft A	Aircraft B	Least Squares Fit	Average of A and B
$n^* = 1.02$	$n^* = 1.01$	$n^* = .92$	$n^* = 1.015$
$C^* = .008$	$C^* = .016$	$C^* = .01$	$C^* = .012$

TABLE 2

C-130 Inspection Data for Selected Locations

(Continued p. 38-60)

Explanation of columns 1-8

- Column 1. Aircraft Identification Number
- Column 2. Location of Rivet Hole (see Fig. 1)
- Column 3. Two Holes at Each Location: (I) Inboard and (O) Outboard
- Column 4. Direction of Crack Growth: (F) Forward and (A) Aft
- Column 5. Crack No. at Rivet Hole. Only First Growing Crack Used
- Column 6. User Command (P) Pacific Air Command, PACAF; (T) Tactical Air Command, TAC
- Column 7. Length of Crack in Hundredths of Inches
- Column 8. Inspection Time in Flight Hours

5807180	73	I	F	5	5	P	6	4904	5807260	89	I	A	1	I	T	19	5629	5807310	74	Q	F	2	P	44	6072
5807180	73	I	F	5	5	P	9	5524	5807260	90	I	A	2	I	T	12	5072	5807310	76	I	A	1	P	12	4921
5807180	73	O	F	3	3	P	25	4904	5807260	90	I	A	2	T	75	5629	5807310	76	I	A	1	P	37	6512	
5807180	73	O	F	3	3	P	50	5082	5807260	91	I	F	2	T	6	4802	5807310	76	I	A	1	P	100	6685	
5807180	73	O	F	3	3	P	62	5213	5807260	91	I	F	2	T	25	5072	5807310	89	I	A	1	P	3	4921	
5807180	74	I	A	2	2	P	3	4904	5807260	91	I	F	2	T	100	5340	5807310	89	I	A	1	P	6	5070	
5807180	74	I	A	2	2	P	6	5334	5807260	91	O	A	1	T	12	4802	5807310	90	O	A	1	P	3	4921	
5807180	74	I	A	2	2	P	9	5822	5807260	91	O	A	1	T	175	5340	5807310	90	O	A	1	P	6	5070	
5807180	74	O	A	1	1	P	3	4904	5807270	75	I	A	1	T	9	5040	5807320	73	I	F	4	T	6	4484	
5807180	74	O	A	1	1	P	6	5334	5807270	75	I	A	1	P	12	5566	5807320	73	I	F	4	T	9	4554	
5807180	74	O	A	1	1	P	12	5822	5807270	89	I	F	1	P	3	4888	5807320	75	I	A	1	T	37	4484	
5807180	75	I	A	2	2	P	25	4904	5807270	89	I	F	1	P	6	5040	5807320	75	I	A	1	T	50	4554	
5807180	75	I	A	2	2	P	50	5082	5807270	89	I	F	1	P	19	5408	5807320	76	I	A	2	T	6	4484	
5807180	75	I	A	2	2	P	62	5213	5807270	89	O	A	2	T	12	4888	5807320	76	I	A	2	T	12	4554	
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6003040	92	I	F	3	P	19	5176	6003090	74	O	A	1	P	12	4516
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6003050	89	O	F	1	P	37	4942	6003090	89	O	F	1	P	56	5067
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6109480	89	0	F	4	P	9	4555	6109500	92	I	A	1	P	37	4772	6109520	92	I	F	2	P	15	4682
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6109710	91	I	A	1	T	75	4439	6126360	75	I	F	4	T	25	4361	6126390	74	O	A	1	1	1	6	P	6	4423
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6377960	89	O	F	1	P	6	4	6	6	4051	6378230	76	O	A	4	T	12	12	4722	6378410	92	I	F	1	T	12	12	3560
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6377960	91	I	F	1	P	19	4	6	19	4946	6378230	89	O	F	1	T	50	50	4373	6378420	73	I	A	1	T	15	15	2781
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6378100	90	I	A	1	T	3	7	6	3	7306	6378250	91	O	A	3	P	19	19	4969	6378490	91	O	A	1	T	25	25	6926
6378100	91	I	A	1	T	6	8	6	6	8306	6378250	92	I	F	2	P	12	12	4123	6378530	91	I	A	1	T	6	6	6052
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6378100	91	I	A	1	T	6	8	6	6	8360	6378250	92	O	A	3	P	12	12	4514	6378570	76	I	F	1	T	19	19	3440

6378570	76	I	F	1	T	25	3577	6378760	89	Q	F	1	T	28	3144	6378770	92	O	F	2	P	9	4502
6378570	91	I	F	2	T	19	2922	6378760	89	O	F	1	T	50	3785	6378770	92	O	F	2	P	12	4724
6378570	31	I	F	2	T	31	3056	6378760	91	I	A	1	T	3	3144	6378770	92	O	F	2	P	19	5058
6378570	91	I	F	2	T	44	3202	6378760	91	I	A	1	T	6	3947	6378800	75	I	A	1	T	6	2880
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6378640	73	O	F	1	T	3	2340	6378760	91	O	F	3	T	3	3144	6378810	75	I	A	1	P	19	4090
6378640	73	O	F	1	T	6	2651	6378760	91	O	F	3	T	6	3340	6378810	75	I	F	2	P	12	3775
6378640	91	I	A	1	T	12	2289	6378760	92	I	F	1	T	3	3144	6378810	75	I	F	2	P	19	4277
6378640	91	I	A	1	T	15	2340	6378760	92	I	F	1	T	6	3947	6378810	75	I	F	2	P	12	4277
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6378680	74	O	A	1	P	3	4699	6378770	73	I	A	2	P	19	5058	6378810	76	O	A	3	P	6	4277
6378680	74	O	A	1	P	6	5210	6378770	73	O	F	1	P	12	3927	6378810	89	I	F	1	P	12	3775
6378680	74	O	A	1	P	12	5484	6378770	73	O	F	1	P	25	4136	6378810	89	I	F	1	P	19	4090
6378680	75	I	F	2	P	9	4221	6378770	73	O	F	1	P	31	5211	6378810	91	I	F	2	P	19	3775
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6378680	75	I	F	2	T	19	5210	6378770	74	O	A	1	P	19	4136	6378810	92	O	F	1	P	12	4090
6378680	75	I	A	3	P	9	4221	6378770	74	O	A	1	P	25	5211	6378810	92	I	F	2	P	3	4090
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6378680	76	I	A	1	P	12	4899	6378770	75	O	A	2	P	12	4724	6378840	92	I	F	1	T	12	2833
6378680	76	O	F	3	P	3	4899	6378770	75	O	A	2	P	25	5211	6378870	75	I	A	2	P	6	3982
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6378680	89	O	A	1	T	12	5210	6378770	76	O	A	1	P	6	3927	6378870	76	O	F	2	P	3	3982
6378680	89	I	A	1	P	12	4899	6378770	76	O	A	1	P	12	4136	6378870	76	O	F	2	P	6	4384
6378680	89	I	A	1	P	25	5720	6378770	76	O	A	2	P	6	3927	6378870	89	I	F	1	P	6	3982
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6378680	90	O	A	1	P	12	4899	6378770	89	I	A	1	P	6	3927	6378870	89	I	F	1	T	12	4715
6378680	90	I	A	3	T	6	5210	6378770	89	I	A	1	P	12	5211	6378870	89	O	A	3	P	6	3982
6378680	90	I	A	3	P	12	5380	6378770	89	I	F	2	P	6	3927	6378870	89	O	A	3	P	9	4123
6378680	91	I	A	3	P	37	4221	6378770	89	I	F	2	P	12	4724	6378870	89	O	F	4	P	6	3982
6378680	91	I	A	3	P	50	4899	6378770	90	O	A	1	P	12	3927	6378870	89	O	F	4	T	9	4687
6378680	91	O	A	5	P	9	4221	6378770	90	O	A	1	P	25	5058	6378870	89	O	F	4	T	12	4715
6378680	91	O	A	5	P	37	6017	6378770	90	I	A	2	P	3	4136	6378870	91	I	A	1	P	12	3665
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6378680	92	I	A	1	P	19	4899	6378770	90	I	A	2	P	12	4821	6378870	91	I	A	1	P	31	4384
6378680	92	O	F	2	T	15	5210	6378770	91	I	A	1	P	6	4821	6378870	91	O	A	3	P	6	3982
6378680	92	O	F	2	T	15	4899	6378770	91	I	A	1	P	19	5058	6378870	91	O	A	3	P	12	4384
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6378760	76	O	F	2	T	3	3144	6378770	92	I	F	1	P	19	3927	6378870	92	O	F	3	P	9	3982
6378760	76	O	F	2	T	6	3947	6378770	92	I	F	1	P	25	4724	6378870	92	O	F	3	T	12	4715
6378760	89	O	F	1	T	12	2966	6378770	92	O	F	2	P	3	3927	6378950	73	O	F	2	T	9	3915

6378950	73	0	F	2	P	19	4488	6378970	73	I	F	2	P	19	4957	6378980	76	I	A	1	Y	6	4387
6378950	73	I	F	3	P	3	4808	6378970	74	I	A	1	P	9	3860	6378980	76	O	A	2	T	6	4387
6378950	73	I	F	3	P	6	4978	6378970	74	I	A	1	P	15	4730	6378980	76	O	A	2	T	9	4679
6378950	73	I	A	4	P	3	4808	6378970	74	O	F	2	P	3	4592	6378990	74	I	A	1	T	3	2788
6378950	73	I	A	4	P	6	4978	6378970	74	O	F	2	P	15	5123	6378990	74	I	A	1	T	6	3360
6378950	74	I	F	2	P	3	4808	6378970	74	O	F	2	P	25	5279	6378990	89	O	F	1	T	19	2788
6378950	74	I	F	2	P	6	5833	6378970	75	I	A	4	P	6	4135	6378990	89	O	F	1	T	44	3125
6378950	74	O	A	1	P	3	4808	6378970	75	I	A	4	P	12	4311	6378990	89	O	F	1	T	50	3360
6378950	74	O	A	1	P	6	5126	6378970	75	I	A	4	P	25	5123	6378990	91	I	A	2	T	12	2788
6378950	75	I	A	3	T	12	3915	6378970	75	O	A	1	P	3	4135	6378990	91	I	A	2	T	44	3125
6378950	75	I	A	3	P	19	4488	6378970	75	O	A	1	P	6	4311	6378990	91	I	A	2	T	50	3360
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6378950	89	O	F	1	T	12	4808	6378970	89	O	F	6	P	12	4592	6378990	73	I	F	2	P	6	3568
6378950	90	O	A	1	P	3	4488	6378970	90	I	A	1	P	3	4311	6378990	73	I	F	2	P	9	4058
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6378950	90	I	A	3	P	3	4808	6378970	90	I	A	1	P	9	5279	6378990	73	I	F	3	P	12	4058
6378950	90	I	A	3	P	6	5730	6378970	90	O	A	2	P	3	4311	6378990	73	O	F	1	P	6	3568
6378950	91	I	A	1	T	6	3915	6378970	91	I	A	3	P	15	5279	6378990	73	O	F	1	P	12	3920
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6378950	91	I	A	1	P	15	5276	6378970	91	O	A	1	P	15	4592	6378990	75	I	A	3	P	22	3568
6378950	91	O	A	3	P	15	5436	6378970	91	O	A	1	P	25	4957	6378990	75	I	A	3	P	25	3920
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6378950	92	I	F	1	P	3	4808	6378970	92	I	F	2	P	12	4135	6378990	75	O	A	1	P	12	3724
6378950	92	I	F	1	P	6	5436	6378970	92	I	F	2	P	15	4592	6378990	75	O	A	1	P	15	3920
6378960	75	O	A	1	T	3	2900	6378970	92	I	F	2	P	19	5404	6378990	76	I	F	2	T	6	3568
6378960	75	O	A	1	T	6	3692	6378970	92	O	A	5	P	3	4135	6378990	76	I	F	2	T	9	4382
6378960	90	I	A	2	T	3	2900	6378970	92	O	A	5	P	25	5279	6378990	76	O	F	1	P	9	3568
6378960	90	I	A	2	T	9	3918	6378980	73	O	F	1	T	6	4387	6378990	76	O	F	1	P	9	3568
6378960	90	O	A	3	T	3	2900	6378980	73	O	F	1	T	12	4679	6378990	89	O	F	1	P	12	3920
6378960	90	O	A	3	T	6	3918	6378980	75	I	A	3	T	3	4276	6378990	89	O	F	1	P	12	3724
6378970	73	O	F	1	P	9	3860	6378980	75	I	A	3	T	6	4387	6378990	89	O	F	1	P	12	3724
6378970	73	O	F	1	P	15	4592	6378980	75	O	A	3	T	6	4387	6378990	89	O	F	1	P	15	3920
6378970	73	O	F	1	P	19	5580	6378980	75	O	A	2	T	6	4276	6378990	89	O	F	1	P	15	4216
6378970	73	I	F	2	P	3	4730	6378980	76	I	A	1	T	12	4582	6378990	89	O	F	1	P	25	4966
6378970	73	I	F	2	P	3	4730	6378980	76	I	A	1	T	3	3825	6378990	90	I	A	1	P	3	3920

6398110	90	I	A	1	P	6	4058	6404960	89	0	F	2	T	12	3961	6404970	92	0	F	3	P	3	4259
6398110	90	0	A	2	P	3	4058	6404960	90	I	A	2	T	6	4242	6404970	92	0	F	3	P	6	4408
6398110	90	0	A	2	P	6	4216	6404960	90	I	A	2	T	12	4700	6404970	92	0	F	3	P	25	5463
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6405700	75	I	A	3	T	50	5340
6405700	75	I	F	4	T	12	4351
6405700	75	I	F	4	T	87	5340
6405700	75	O	A	1	T	6	4351
6405700	75	O	A	1	T	12	5340
6405700	89	I	F	2	T	3	4351
6405700	89	I	F	2	T	12	5340
6405700	89	O	A	1	T	9	4351
6405700	89	O	A	1	T	25	5340
6405700	90	I	A	2	T	3	4351
6405700	90	I	A	2	T	12	4895
6405700	90	O	A	1	T	12	4895
6405700	91	I	A	1	T	6	4351
6405700	91	O	A	1	T	12	4701
6405700	91	O	A	3	T	6	4701
6405700	92	O	F	1	T	3	4351
6405700	92	O	F	1	T	12	4895
6405700	92	I	F	3	T	6	4895
6405700	92	I	F	3	T	25	5340
6405710	74	O	A	1	T	6	4890
6405710	74	O	A	1	T	9	5030
6417681	73	O	F	1	T	19	3578
6417681	73	O	F	1	T	25	3905

TABLE 3

Crack Propagation Parameters for Constant n^*

Location	\bar{n}^*	mean value of			coefficient of variation of			No. of Data Pts.	
		\bar{C}^*	\bar{t}_i hrs.	\bar{a}_0 in.	n^*	C^*	t_i		a_0
76 92 TAC	1.500	0.00537	3385.	0.00527	0.0	0.508	0.329	0.750	130
76 92 PACAF	1.500	0.00788	4178.	0.00240	0.0	0.655	0.291	0.596	188
75 91 TAC	1.500	0.00623	3383.	0.00457	0.0	0.640	0.352	0.934	167
75 91 PACAF	1.500	0.00525	3750	0.00498	0.0	0.623	0.387	0.650	268
74 90 TAC	1.500	0.00469	3637.	0.00550	0.0	0.614	0.338	0.433	75
74 90 PACAF	1.500	0.00660	4394.	0.00273	0.0	0.330	0.296	0.405	151
73 89 TAC	1.500	0.00562	3446.	0.00483	0.0	0.774	0.336	0.606	101
73 89 PACAF	1.500	0.00486	3930.	0.00489	0.0	0.671	0.331	0.506	176

TABLE 4

Statistical Parameters for Time to Crack Initiation
Two Parameter Weibull and Johnson S_B Distributions

Location	Two Parameter Weibull Distribution			Johnson S_B Distribution					
	α	β		β_1	β_2	ϵ	λ	η	γ
76 92 TAC	3.796	3760.		0.643	5.043	0.0	91,000.	2.980	9.821
76 92 PACAF	4.125	4602.		1.105	5.702	0.0	51,080.	4.007	9.879
75 91 TAC	3.545	3770.		0.917	4.450	0.0	82,950.	3.033	9.803
75 91 PACAF	3.098	4194.		1.348	4.876	0.0	45,210.	2.140	5.336
74 90 TAC	3.553	4039.		1.408	4.298	0.0	108,500.	2.514	10.160
74 90 PACAF	4.044	4844.		2.054	5.567	0.0	89,620.	3.086	9.413
73 89 TAC	3.571	3826.		1.243	4.705	0.0	66,150.	2.841	8.534
73 89 PACAF	3.629	4359.		1.136	4.971	0.0	29,460.	2.120	4.147

TABLE 5

Statistical Parameters for Fitting Time to Crack Initiation
Three Parameter Weibull Distribution and Convolution

Location	Three Parameter Weibull Distribution			Convolution			λ
	α	β	x_0	α	β	x_0	
76 92 TAC	9.5	3500	-2000	9.5	3900	-2000	.003
76 92 PACAF	10.5	4250	-1900	10.5	5000	-1900	.003
75 91 TAC	9.5	3400	-2000	9.5	3500	-2000	.016
75 91 PACAF	6.5	3750	-800	6.5	4000	-800	.014
74 90 TAC	9.5	3500	-1000	9.5	3550	-1000	.014
74 90 PACAF	10.5	4300	-1500	10.5	4500	-1500	.014
73 89 TAC	9.5	3250	-1000	9.5	3450	-1000	.012
73 89 PACAF	6.5	3950	-500	6.5	4200	-500	.012

TABLE 6

Statistical Parameters for Initial Flaw Sizes, a_0 , in.

Location	Mean	Standard Deviation	Johnson S_B Distribution						
			β_1	β_2	ϵ	λ in.	η	γ	
76 92	TAC	0.00527	0.00396	20.2	33.8	0.0	0.03	2.364	4.048
76 92	PACAF	0.00240	0.00143	12.0	18.0	0.0	0.03	2.473	6.579
75 91	TAC	0.00457	0.00428	2.8	7.6	0.0	0.03	2.010	3.894
75 91	PACAF	0.00498	0.00323	8.8	16.6	0.0	0.03	1.767	3.314
74 90	TAC	0.00550	0.00238	5.5	14.6	0.0	0.03	2.034	3.382
74 90	PACAF	0.00273	0.00110	10.0	26.6	0.0	0.03	2.046	5.167
73 89	TAC	0.00483	0.00293	21.8	32.4	0.0	0.03	2.013	3.781
73 89	PACAF	0.00489	0.00248	8.8	16.7	0.0	0.03	2.056	3.672
Over all		0.00431	0.00001	37.9	71.7	0.0	0.03	1.500	2.900

TABLE 7

Normalizing Constants and Percentage of Detected and Undetected Crack Initiation Times

Inspection Time t hours	k_3	% Detected	k_4	% Undetected
2500	16.60	6.02	1.06	93.98
4000	1.60	62.56	2.67	37.44
5000	1.01	98.56	69.34	1.44

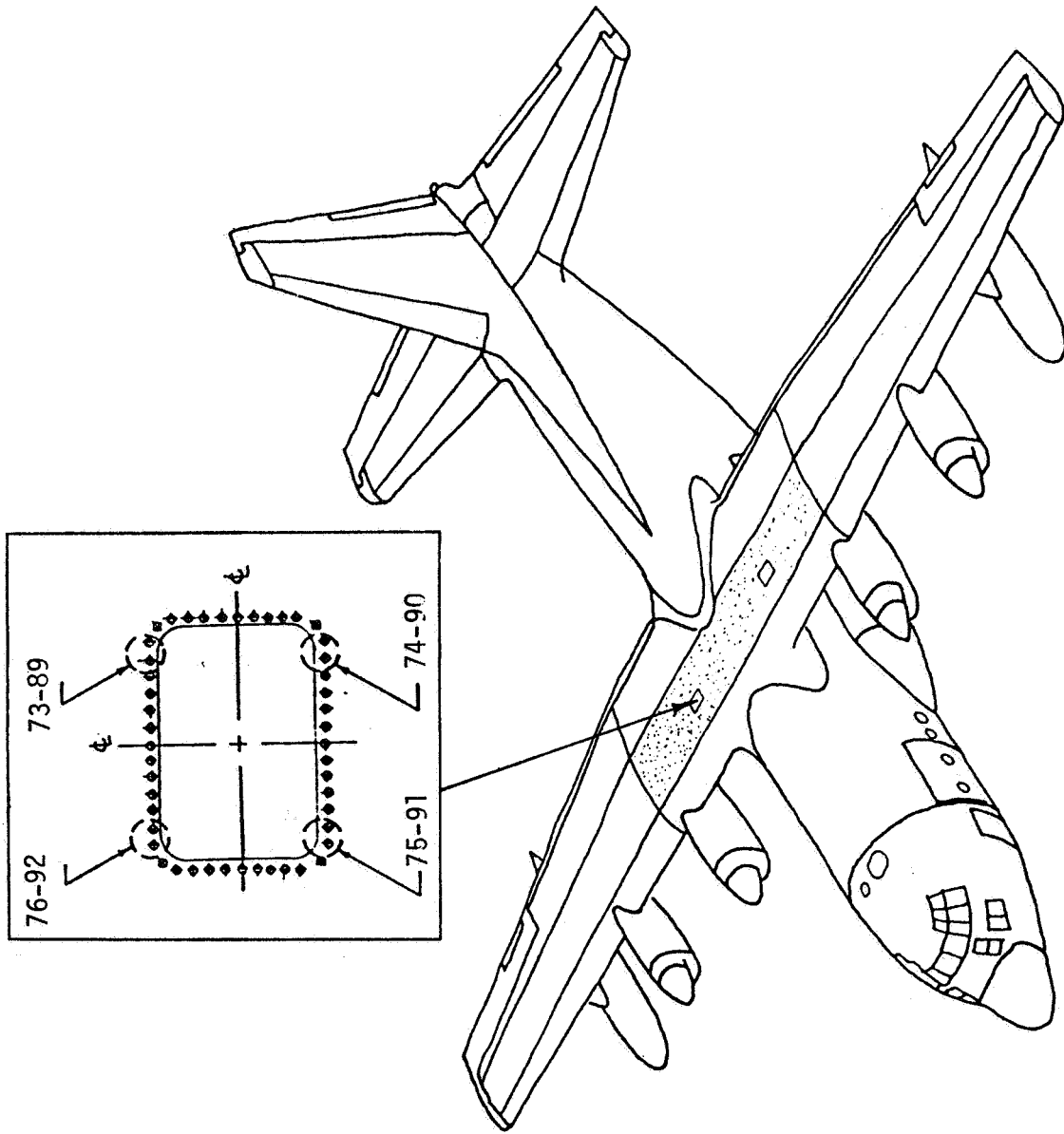


Fig. 1. Location of Investigated Cracks on C-130 Aircraft.

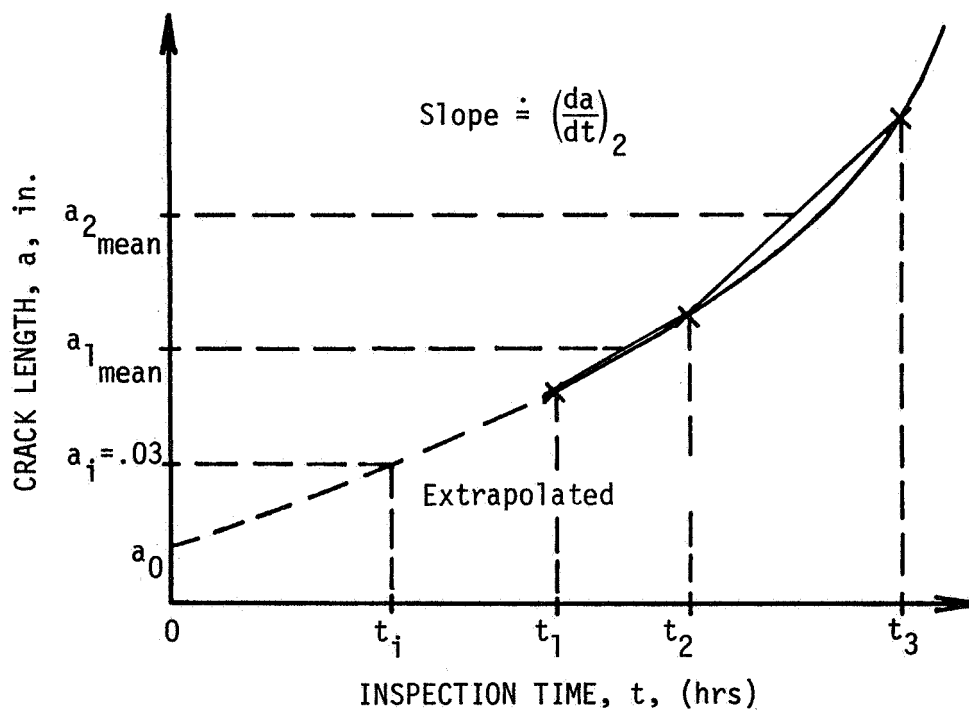


Fig. 2. Extrapolation of Inspection Data.

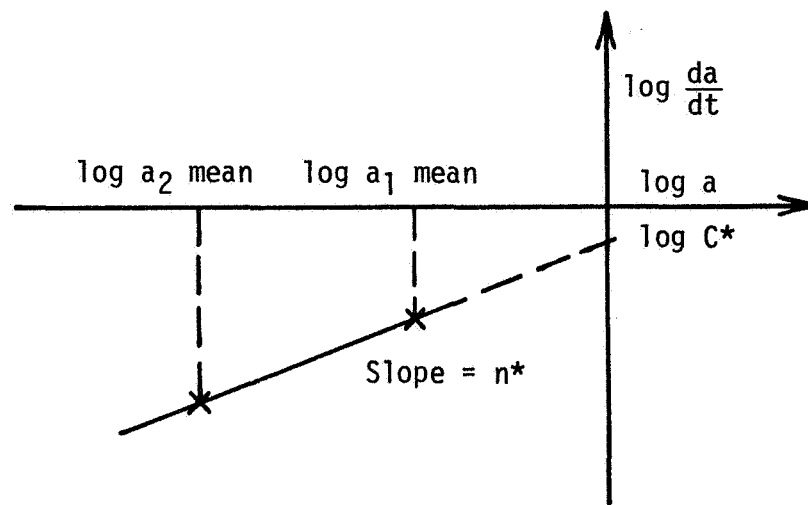


Fig. 3. Determination of Crack Growth Parameters. Variable n^* Method.

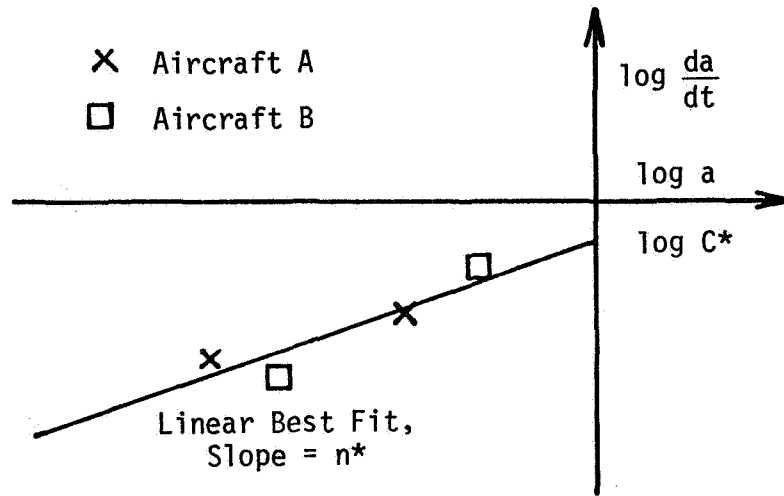


Fig. 4. Determination of Crack Growth Parameters. Linear Regression Method.

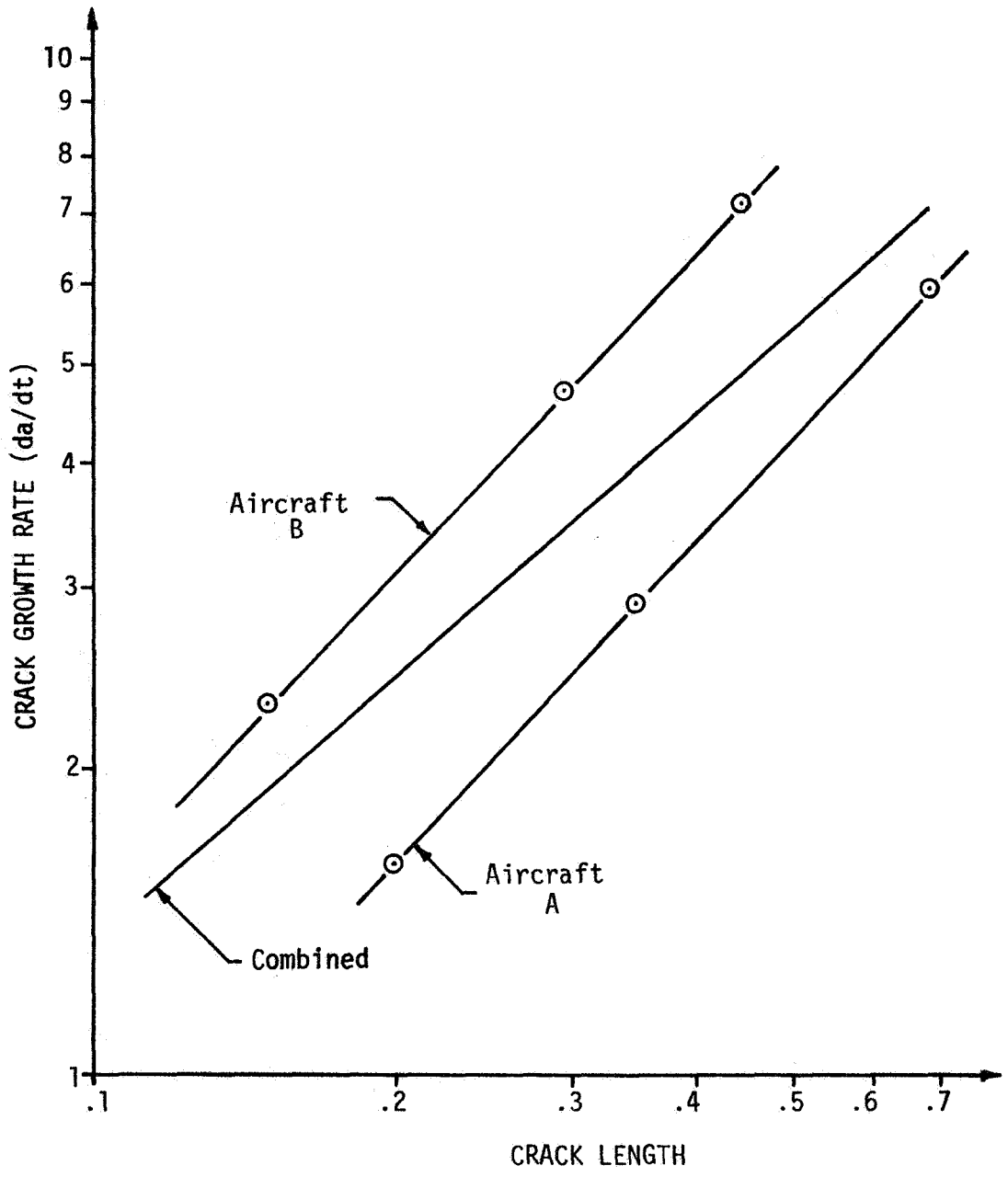


Fig. 5. Two Ideally Growing Cracks.

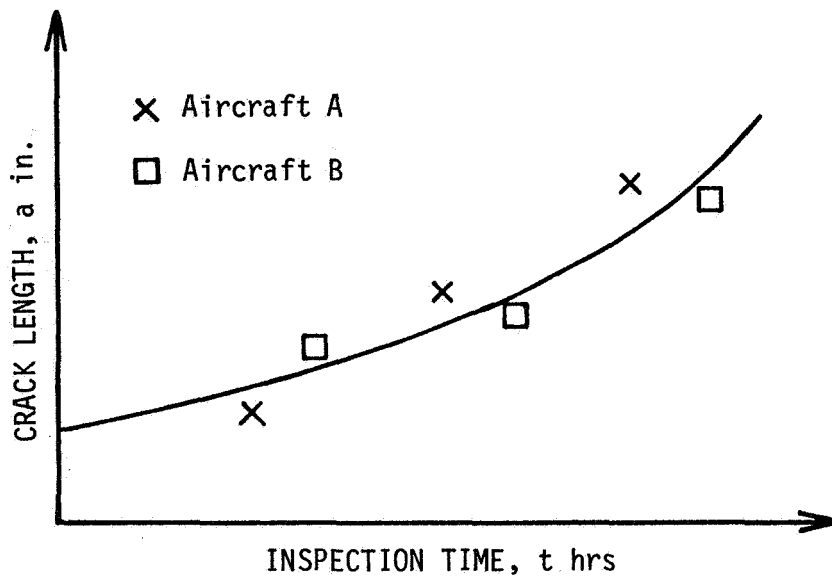


Fig. 6. Determination of Crack Growth Parameters. Nonlinear Regression Method.

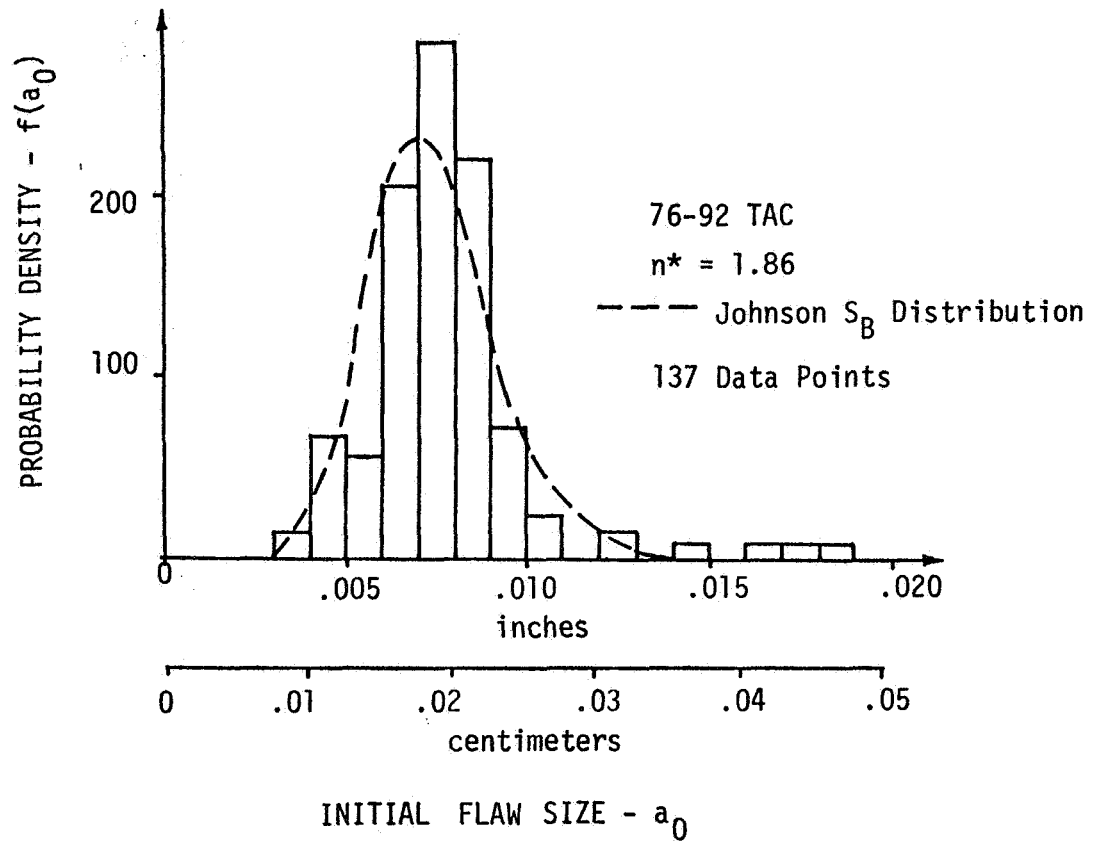


Fig. 7. Histogram of Initial Flaw Size Using a Larger n^* .

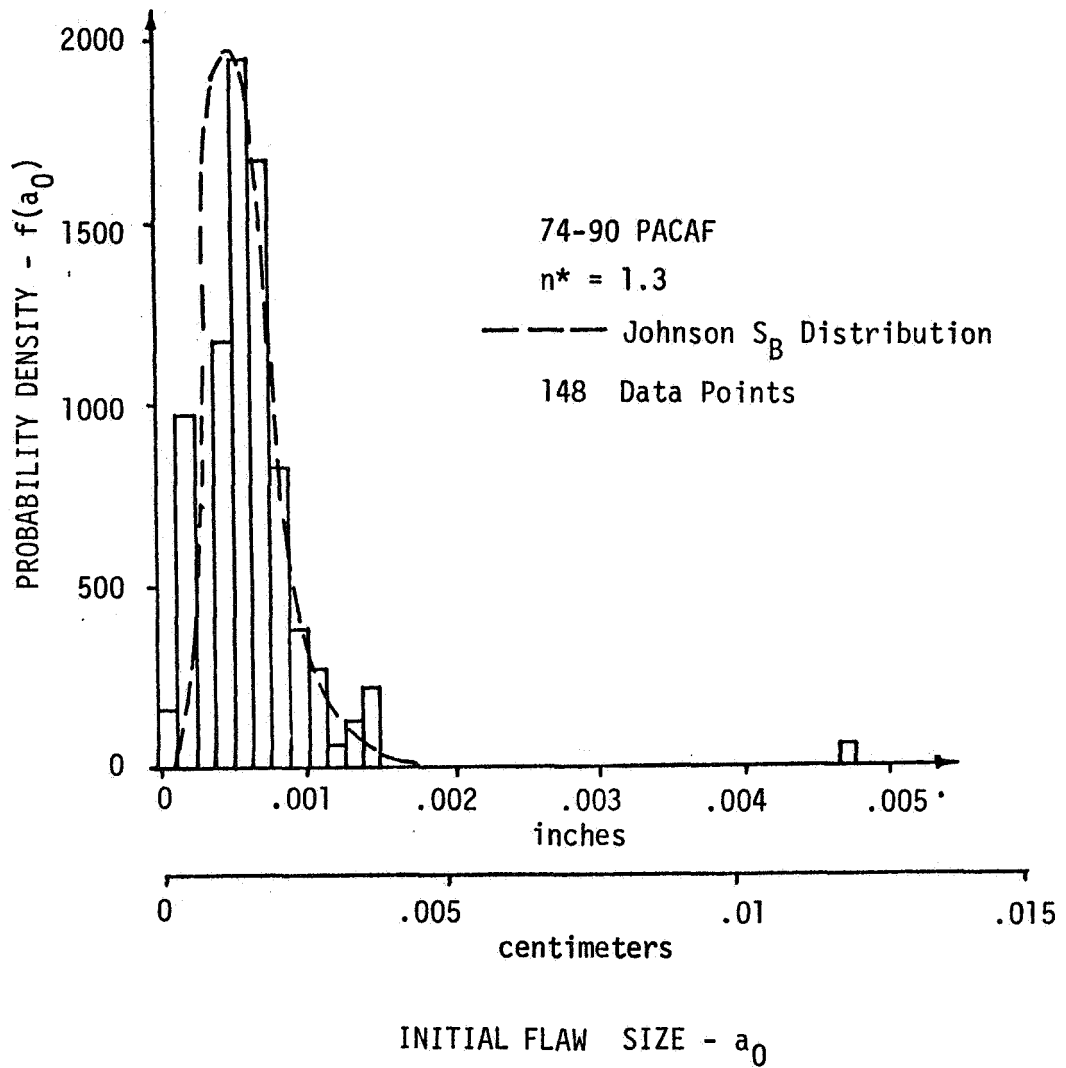


Fig. 8. Histogram of Initial Flaw Size Using a Smaller n^* .

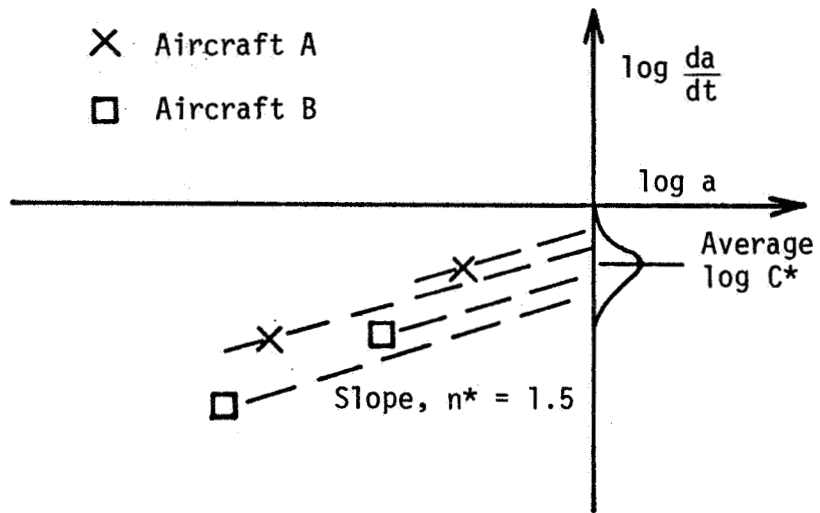


Fig. 9. Determination of Crack Growth Parameters. Constant n^* Method.

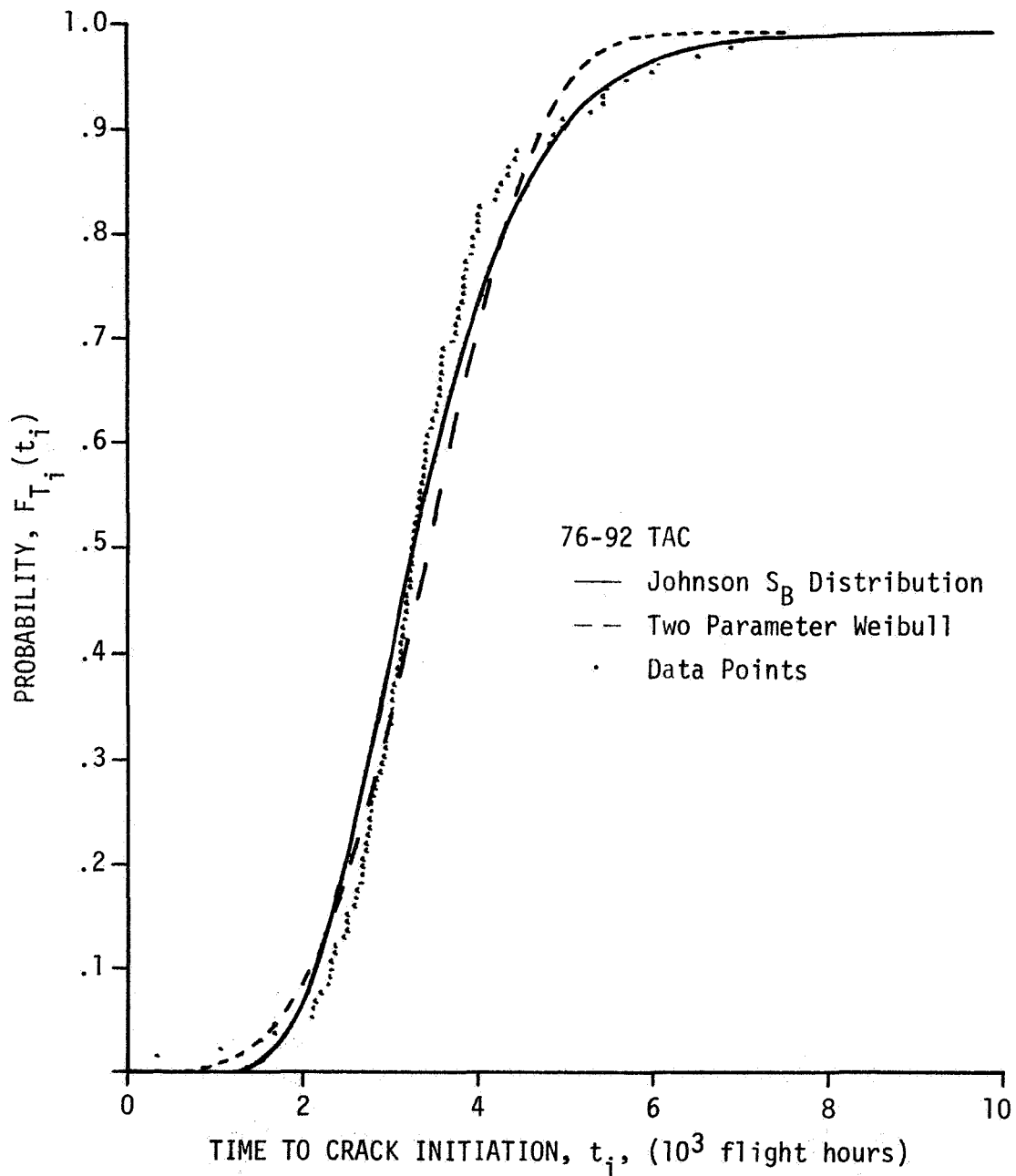


Fig. 10. Distribution of Crack Initiation Times. Two Parameter Weibull and Johnson S_B Distributions.

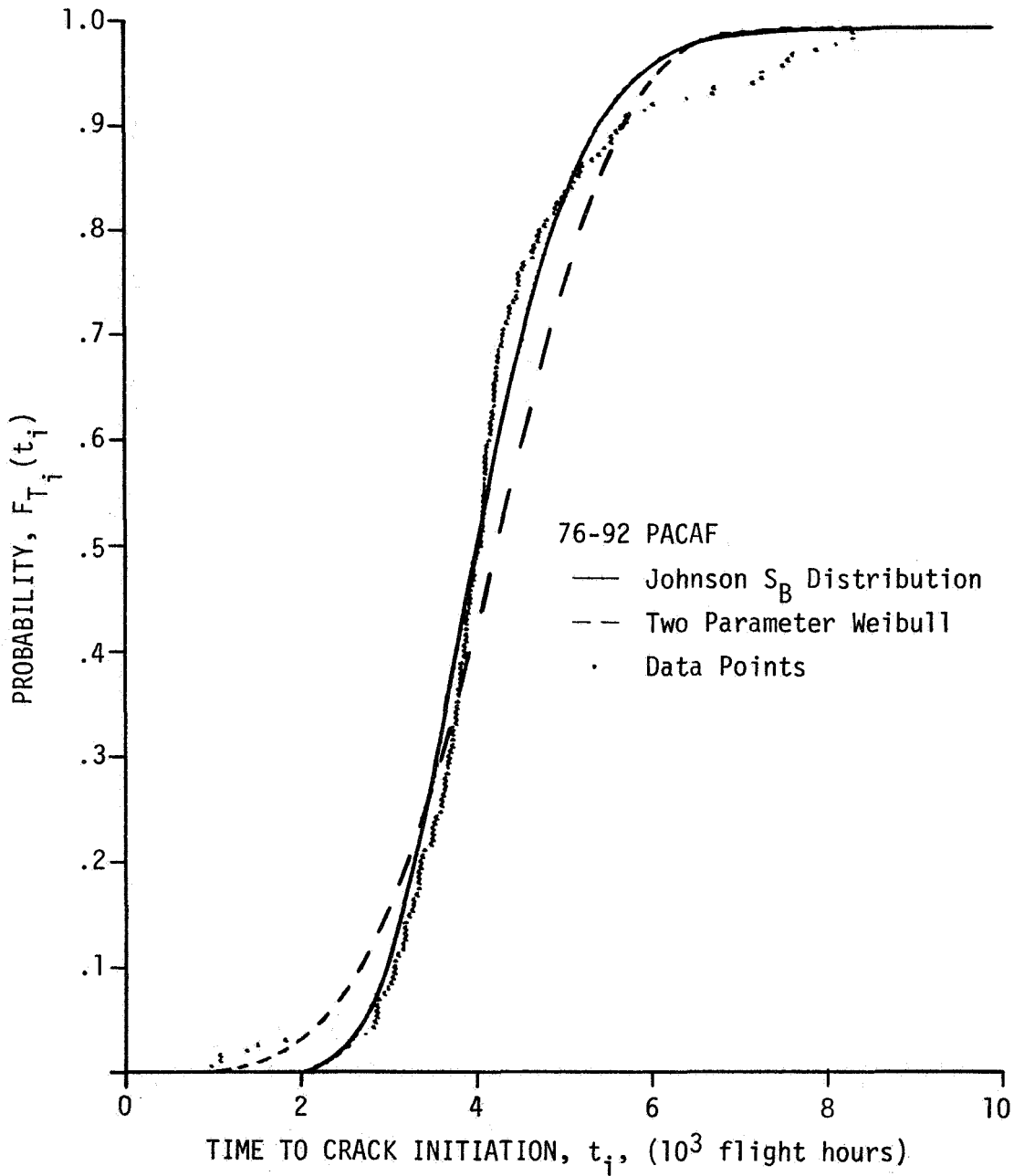


Fig. 11. Distribution of Crack Initiation Times. Two Parameter Weibull and Johnson S_B Distributions.

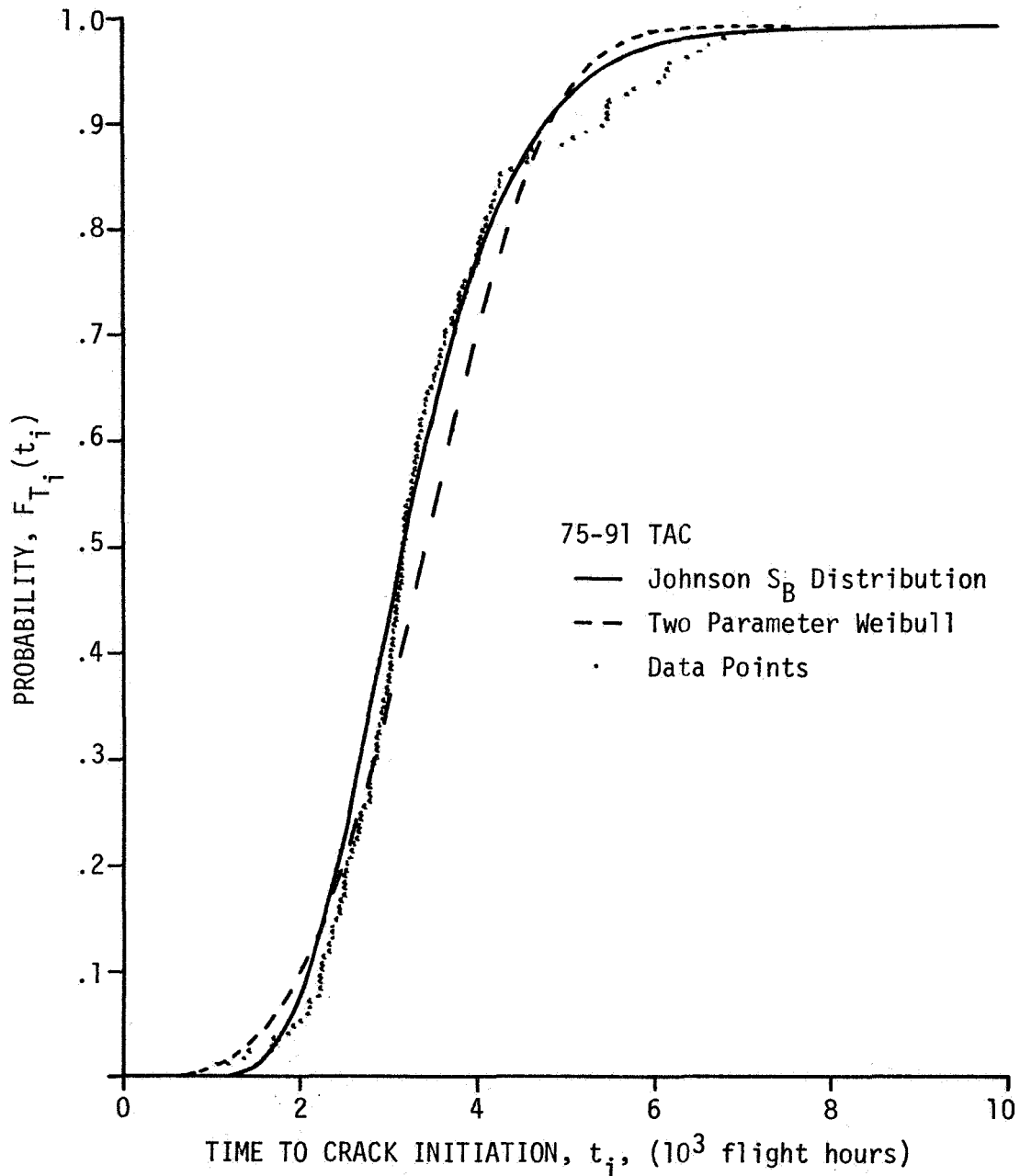


Fig. 12. Distribution of Crack Initiation Times. Two Parameter Weibull and Johnson S_B Distributions.

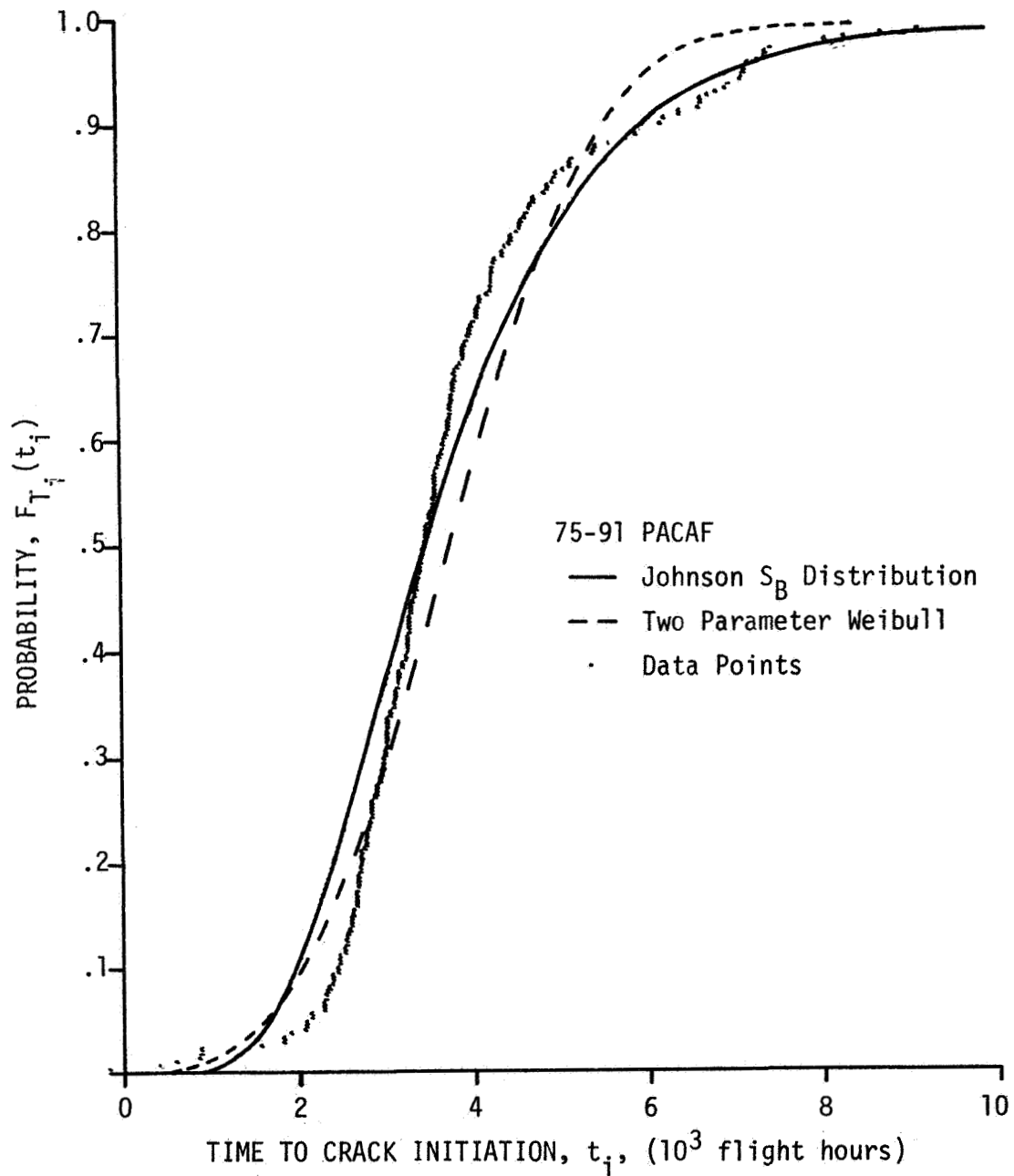


Fig. 13. Distribution of Crack Initiation Times. Two Parameter Weibull and Johnson S_B Distributions.

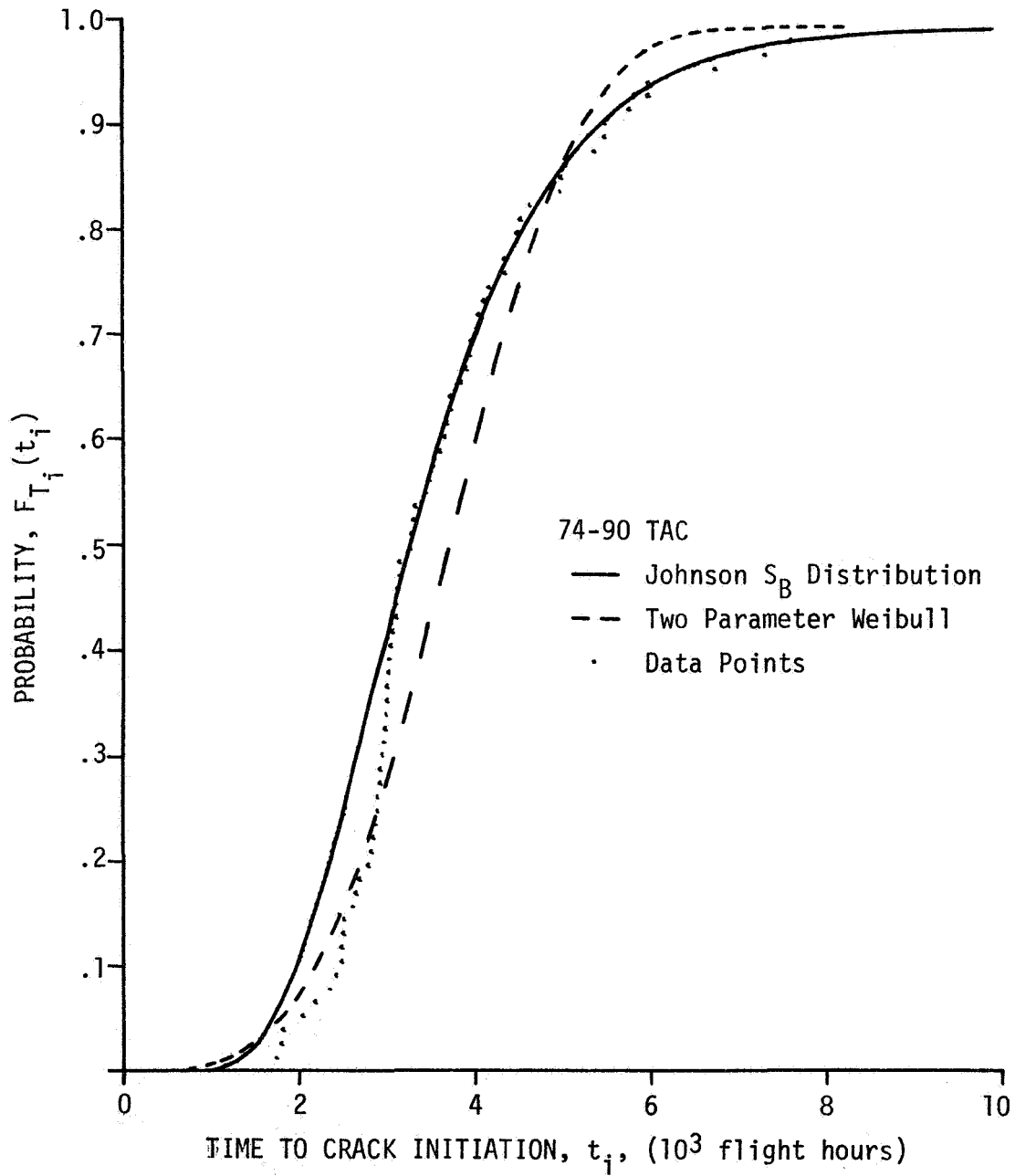


Fig. 14. Distribution of Crack Initiation Times. Two Parameter Weibull and Johnson S_B Distributions.

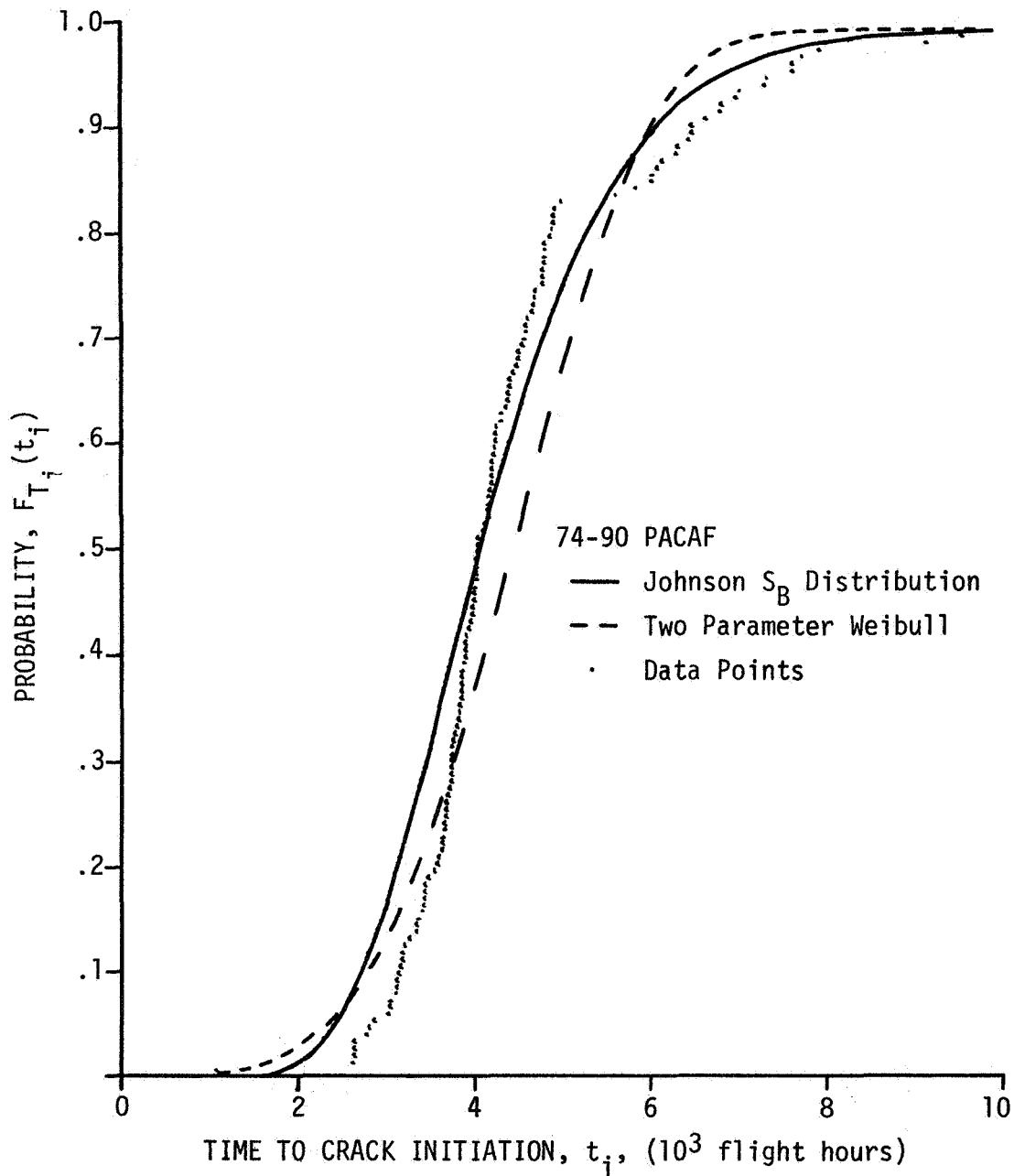


Fig. 15. Distribution of Crack Initiation Times. Two Parameter Weibull and Johnson S_B Distributions.

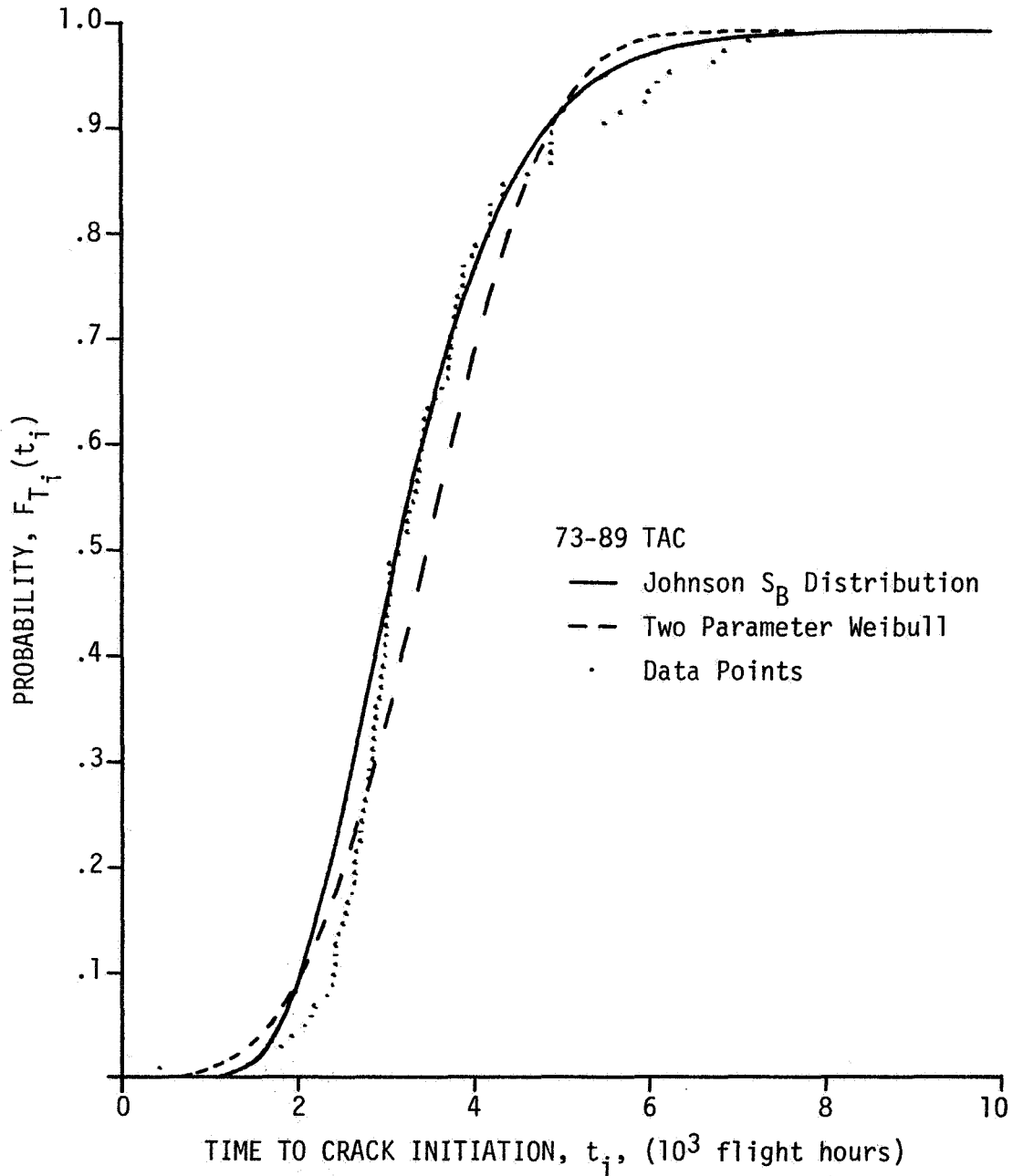


Fig. 16. Distribution of Crack Initiation Times. Two Parameter Weibull and Johnson S_B Distributions.

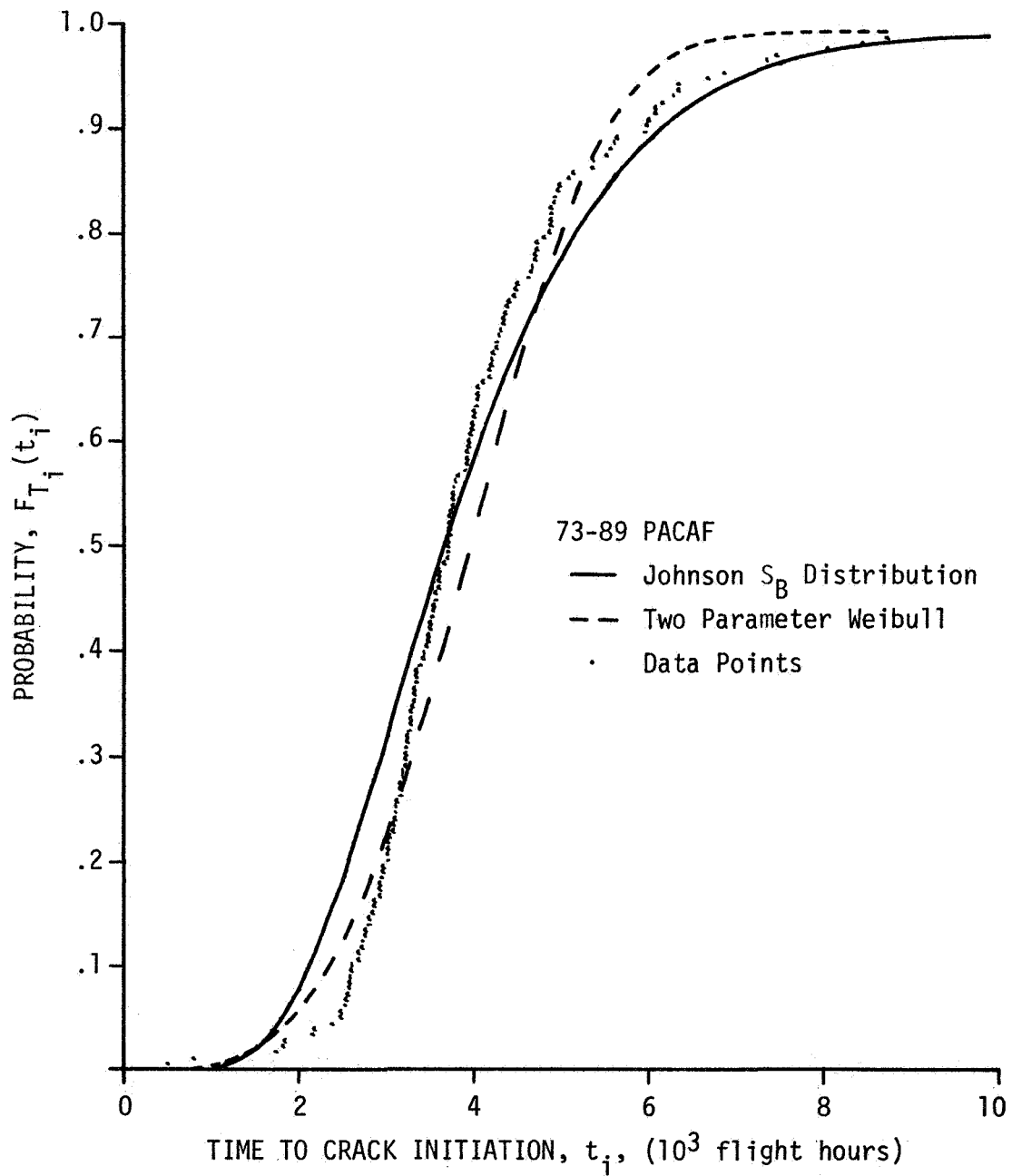


Fig. 17. Distribution of Time to Crack Initiation. Two Parameter Weibull and Johnson S_B Distributions.

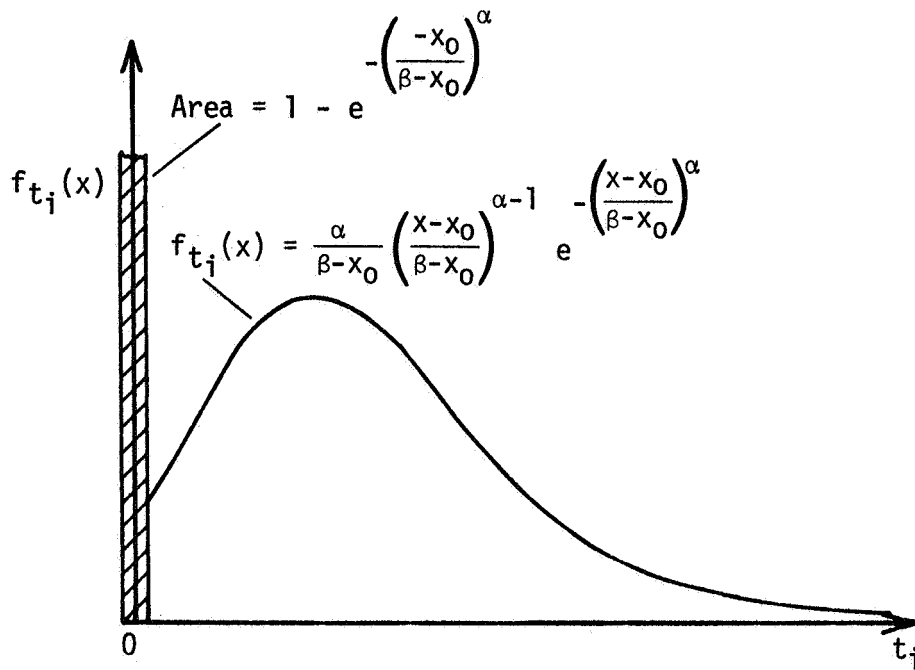


Fig. 18. Three Parameter Weibull Density Function of Crack Initiation Times.

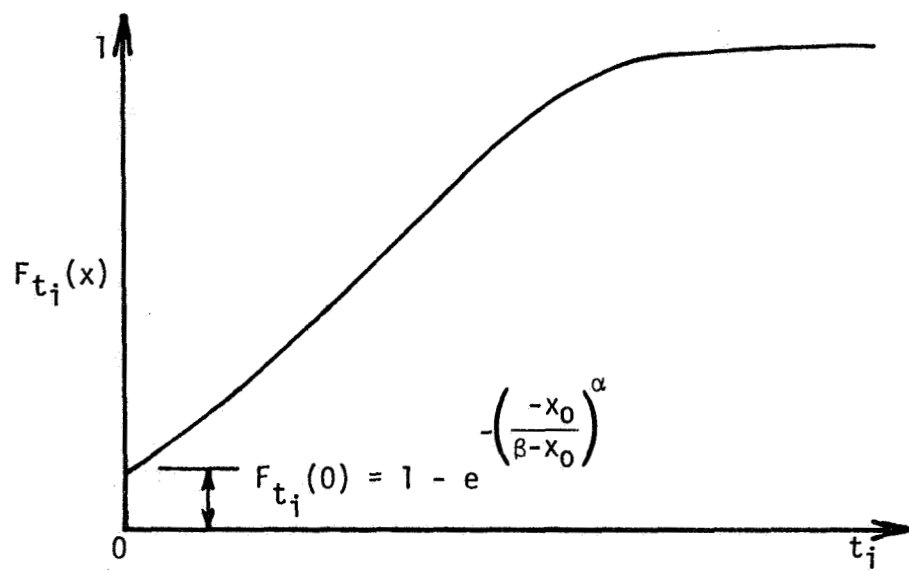


Fig. 19. Three Parameter Weibull Distribution Function of Crack Initiation Times.

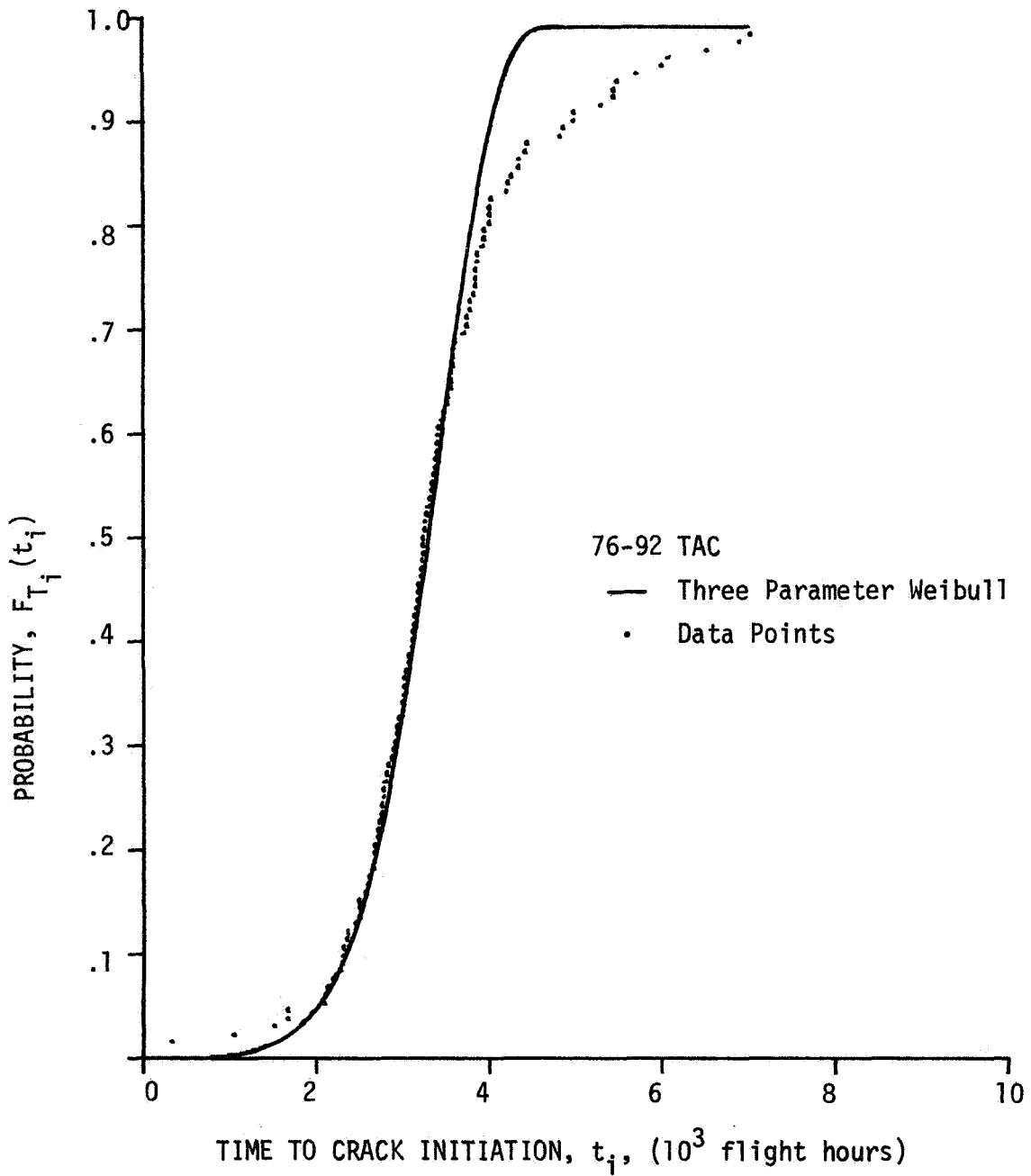


Fig. 20. Distribution of Crack Initiation Times.

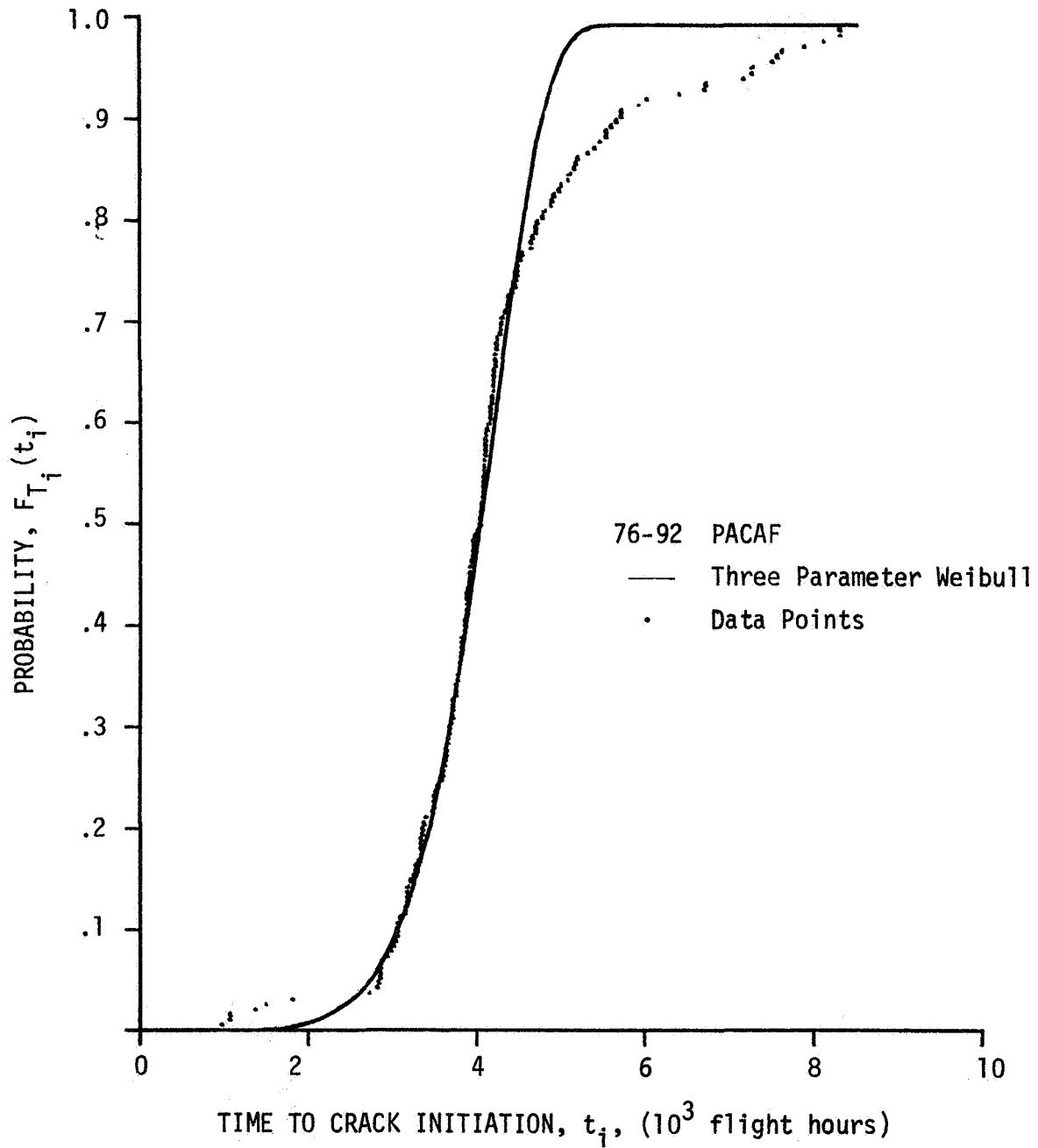


Fig. 21. Distribution of Crack Initiation Time.

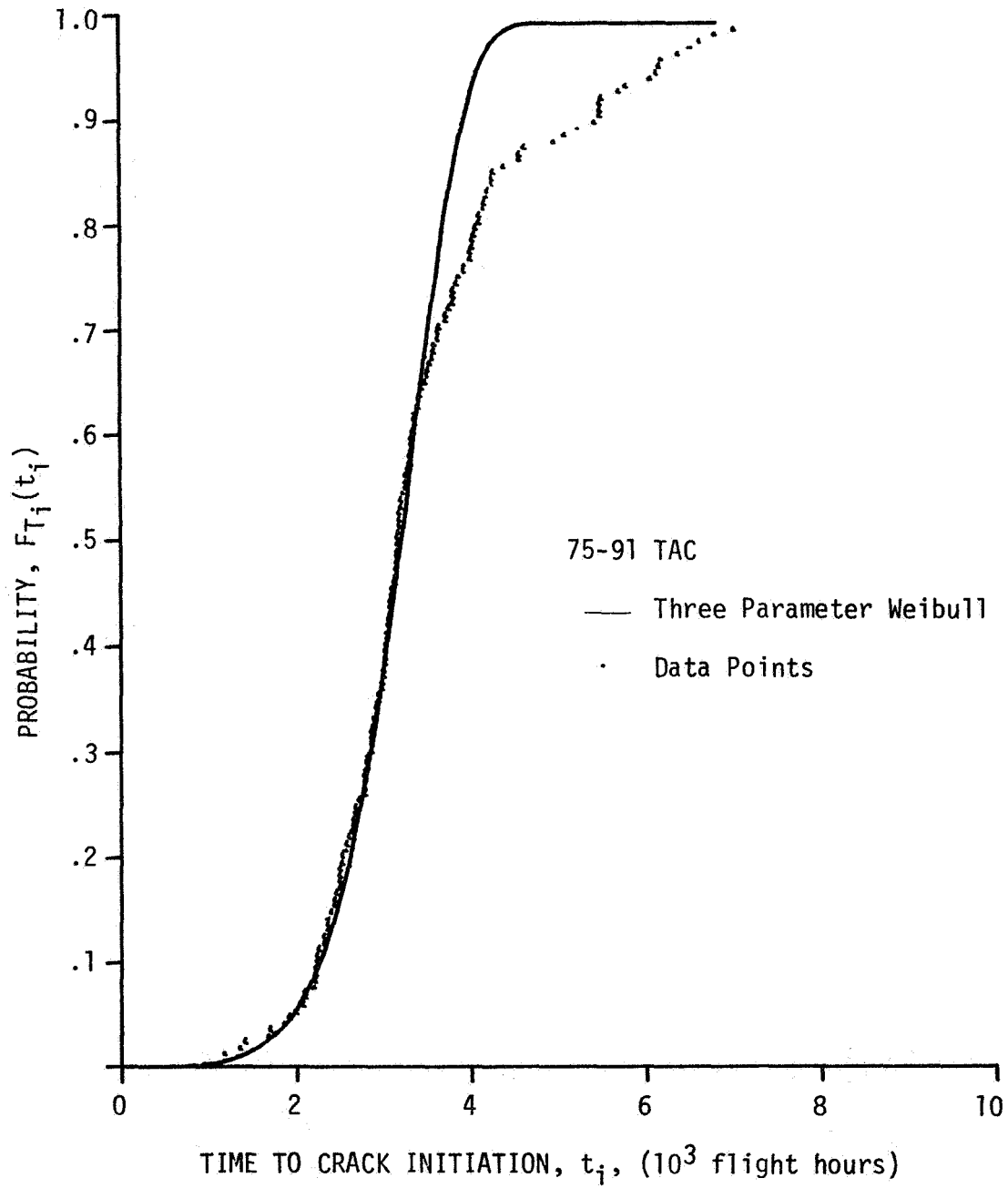


Fig. 22. Distribution of Crack Initiation Times.

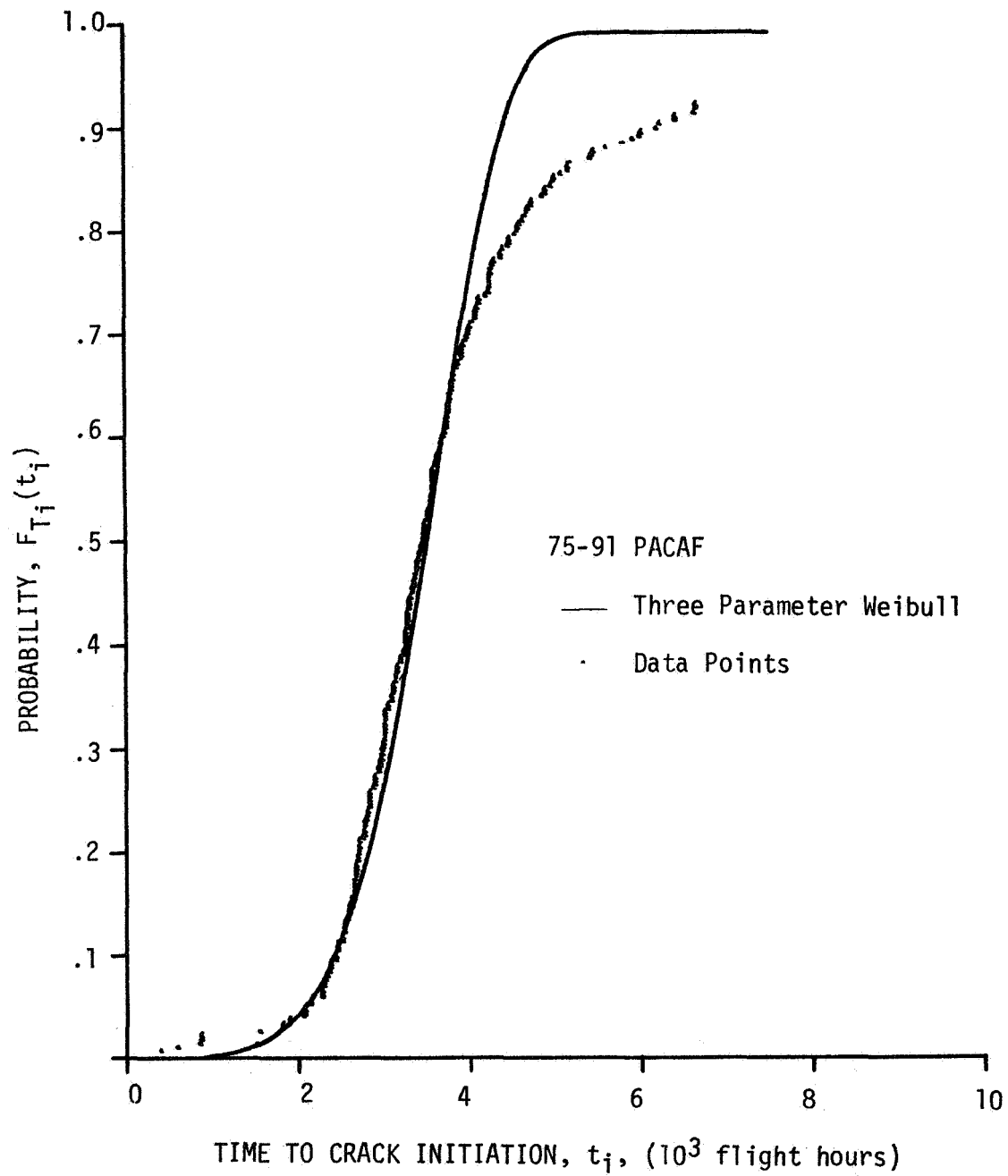


Fig. 23. Distribution of Crack Initiation Times.

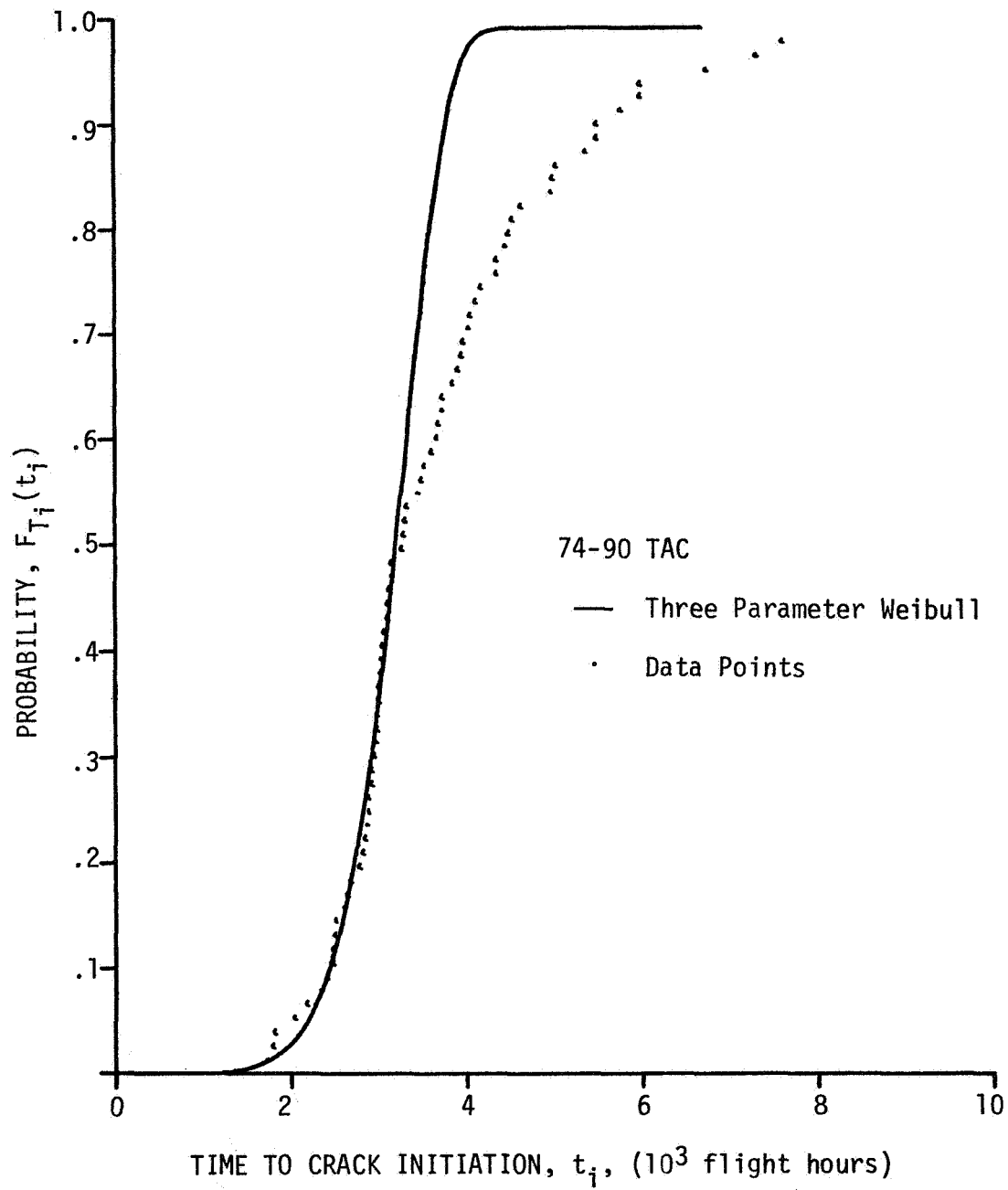


Fig. 24. Distribution of Crack Initiation Times.

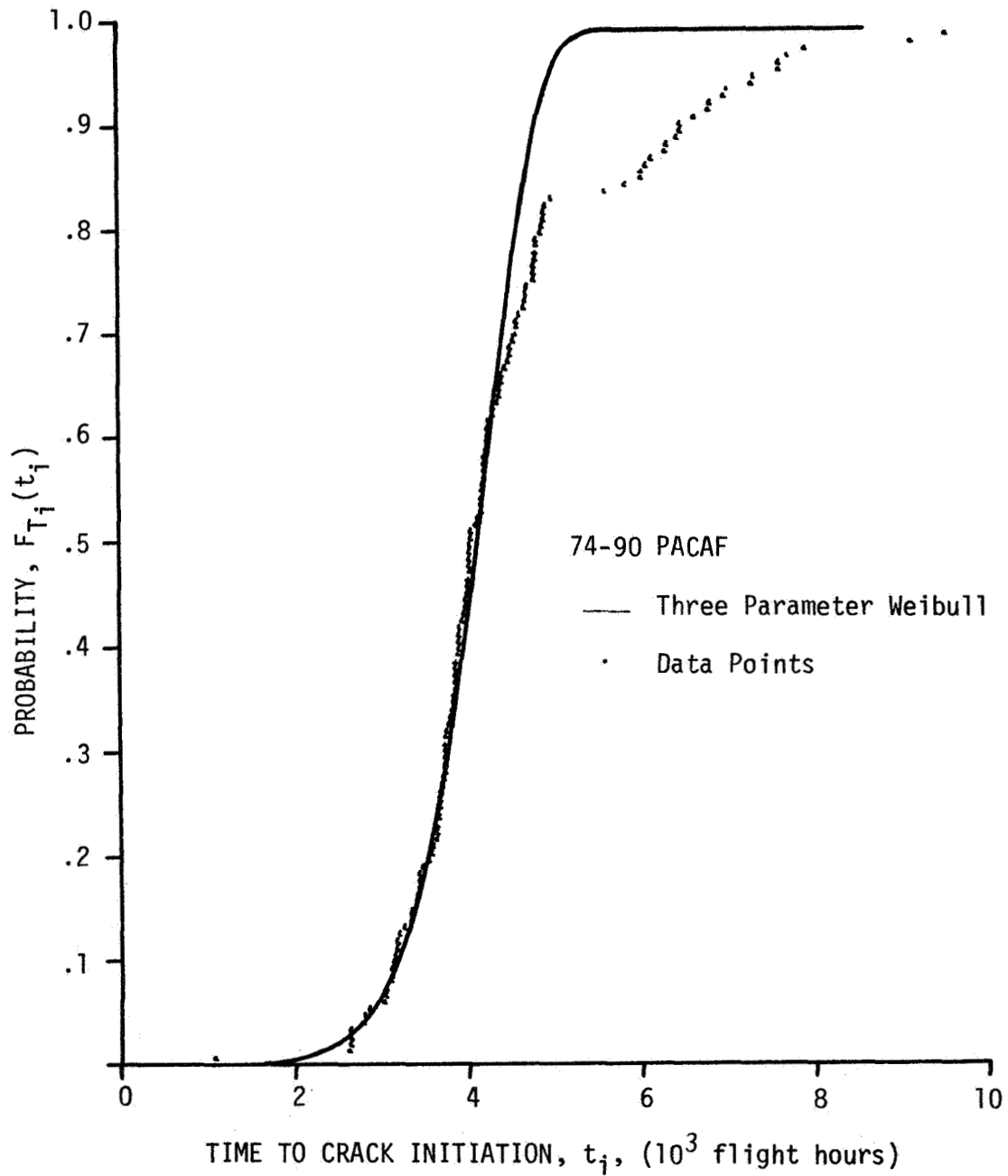


Fig. 25. Distribution of Crack Initiation Times.

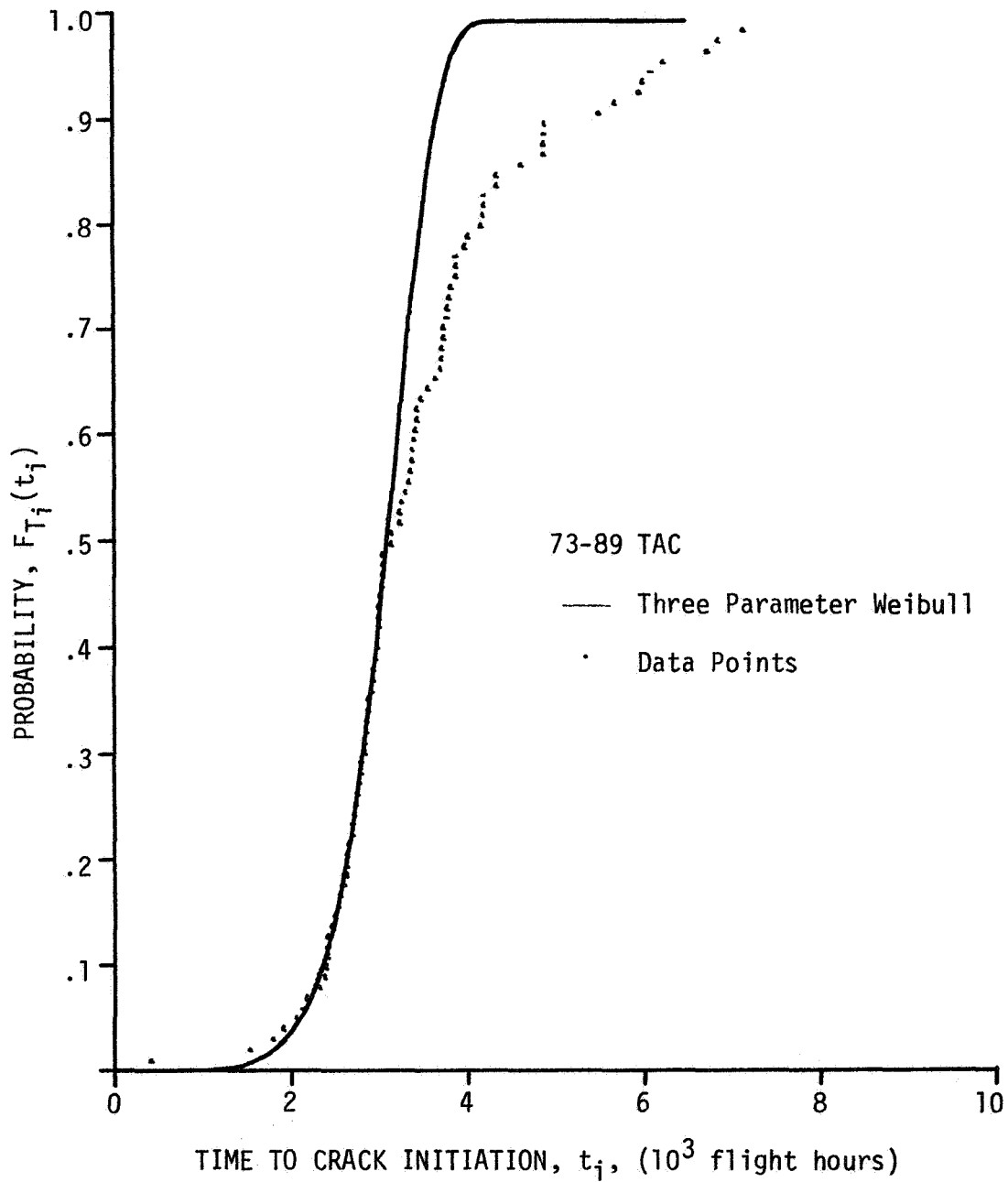


Fig. 26. Distribution of Crack Initiation Times.

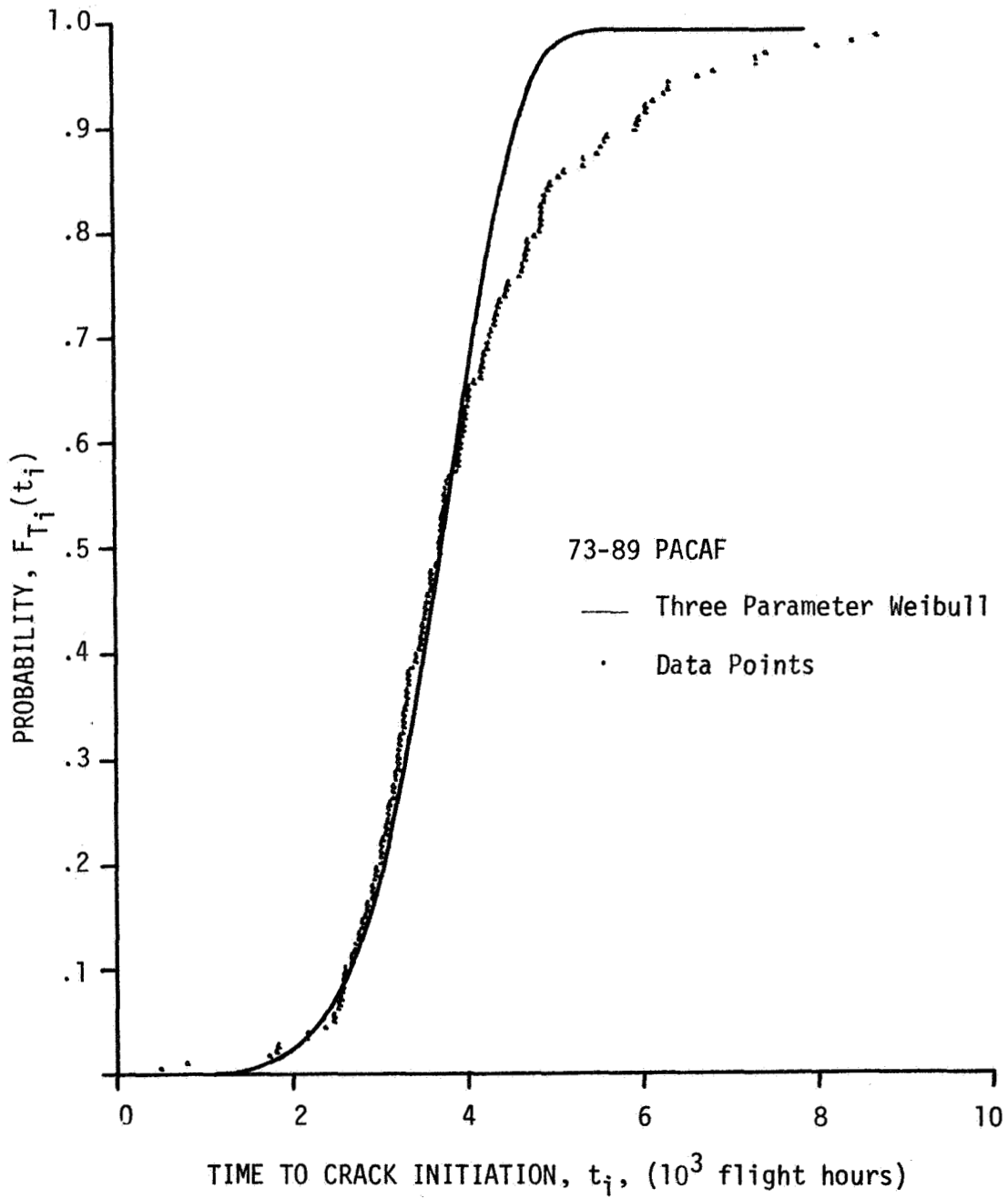


Fig. 27. Distribution of Crack Initiation Times.

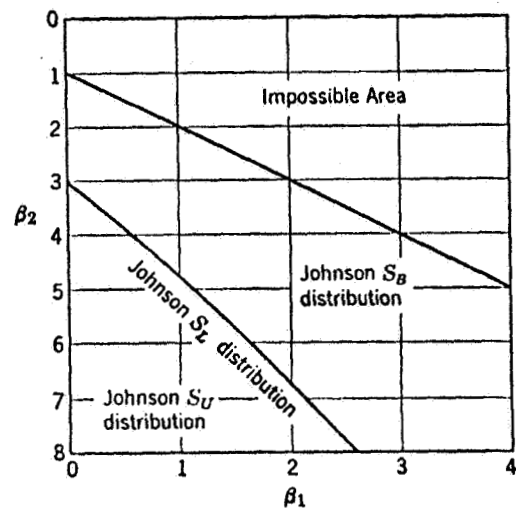


Fig. 28. Johnson Distribution Applicability Chart.

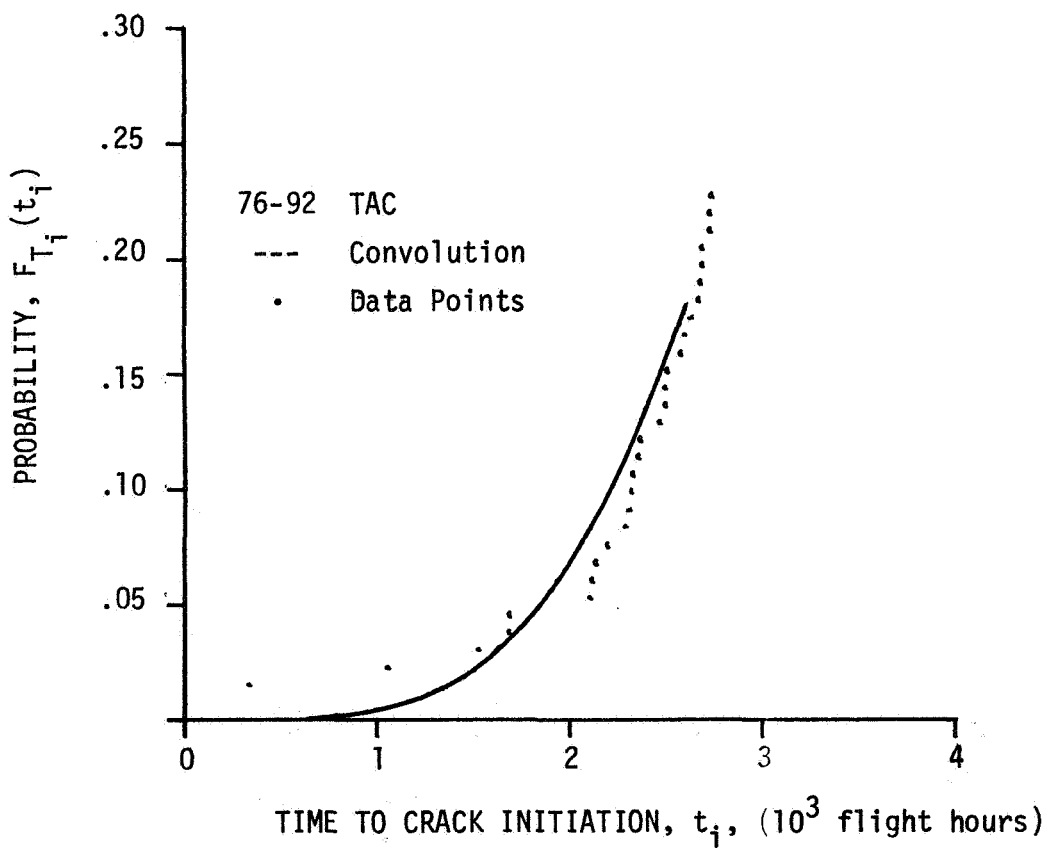


Fig. 29. Lower Tail of Crack Initiation Time Distribution

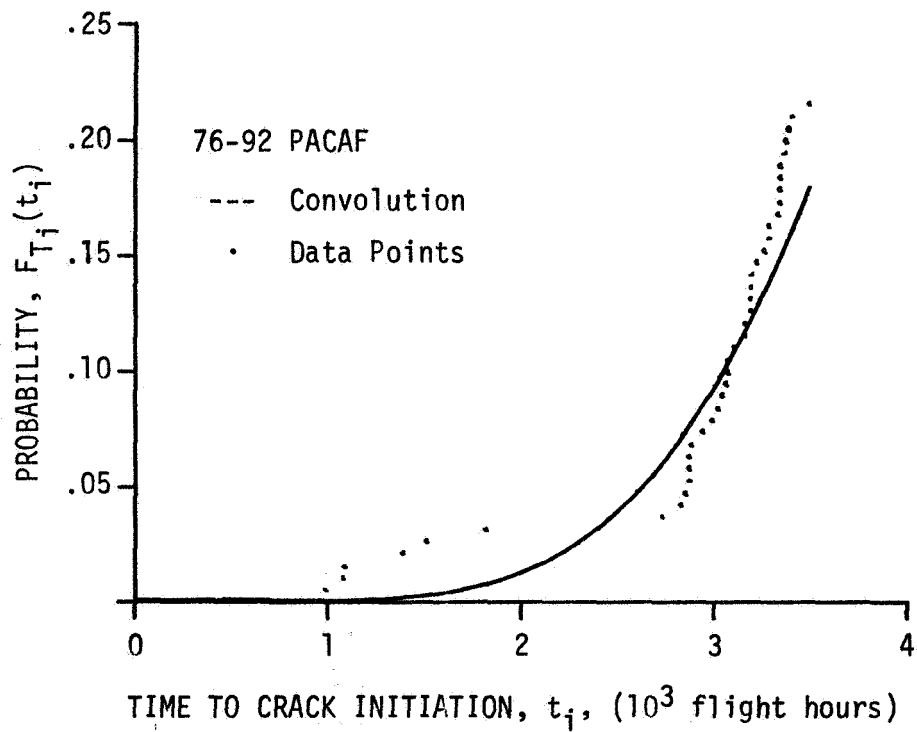


Fig. 30. Lower Tail of Crack Initiation Time Distribution.

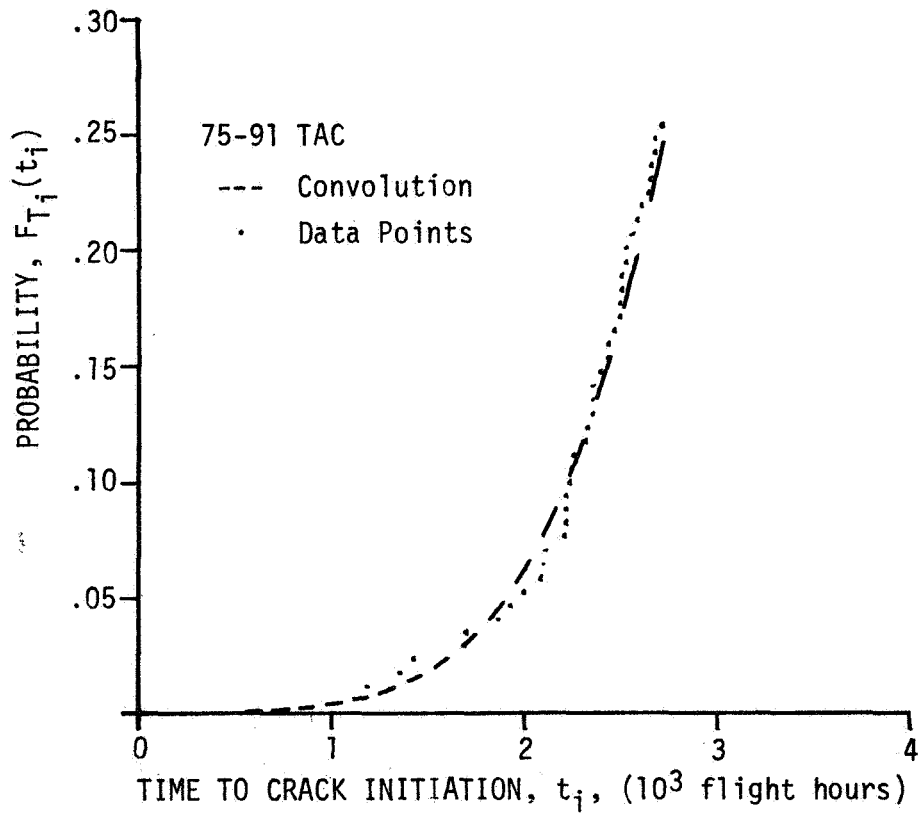


Fig. 31. Lower Tail of Crack Initiation Time Distribution.

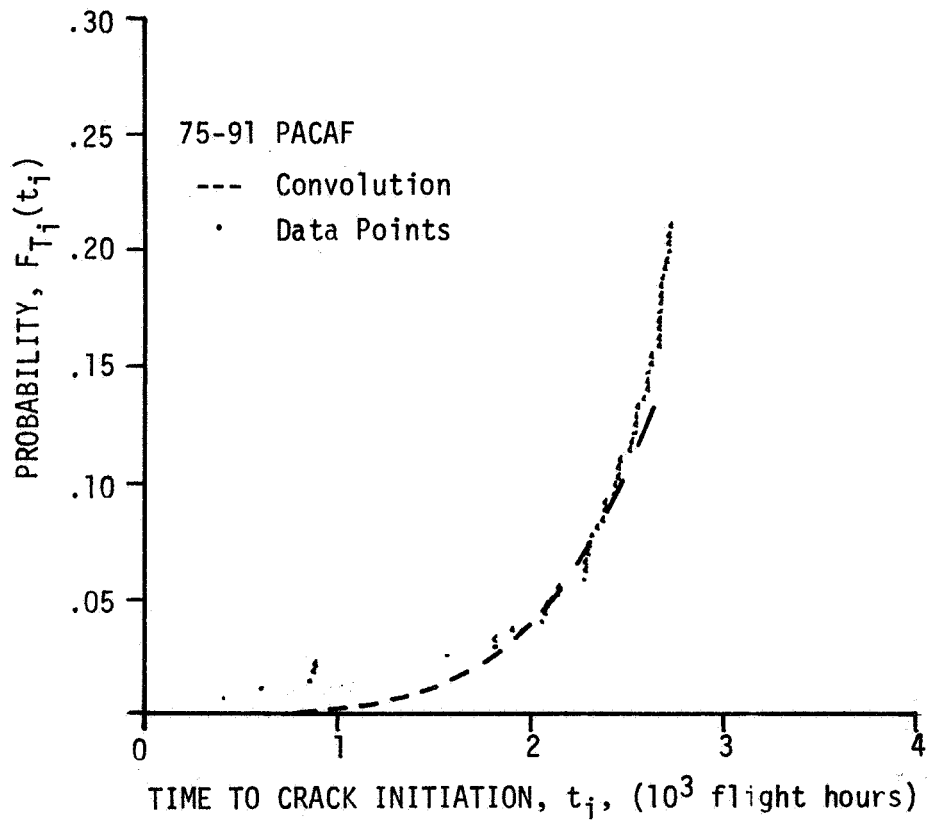


Fig. 32. Lower Tail of Crack Initiation Time Distribution.

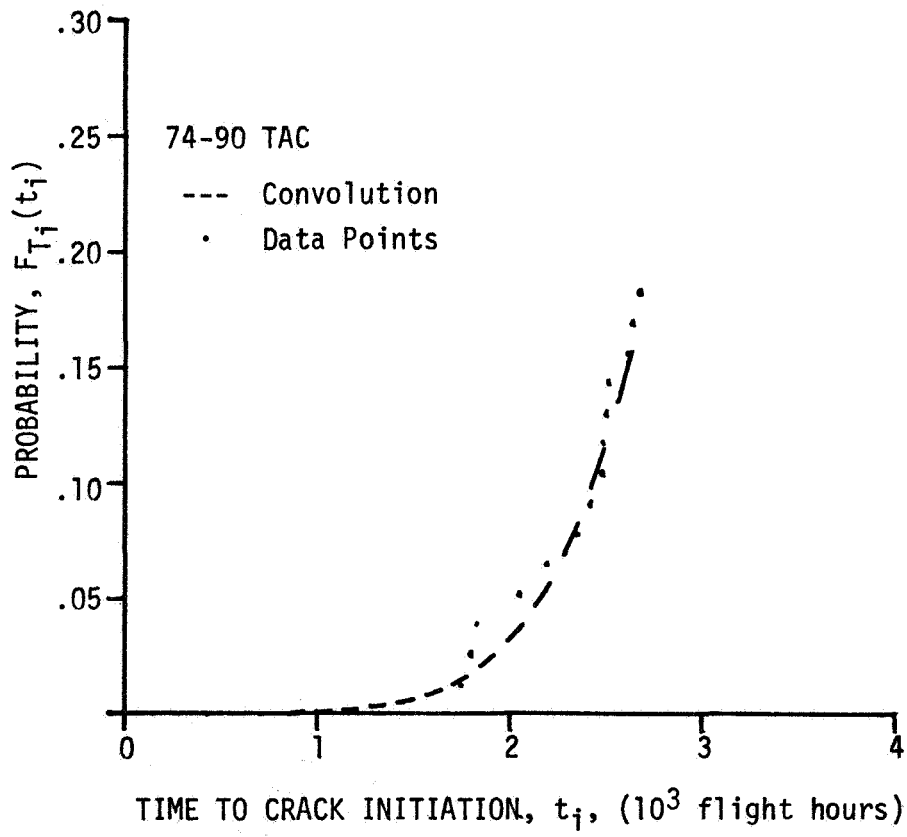


Fig. 33. Lower Tail of Crack Initiation Time Distribution.

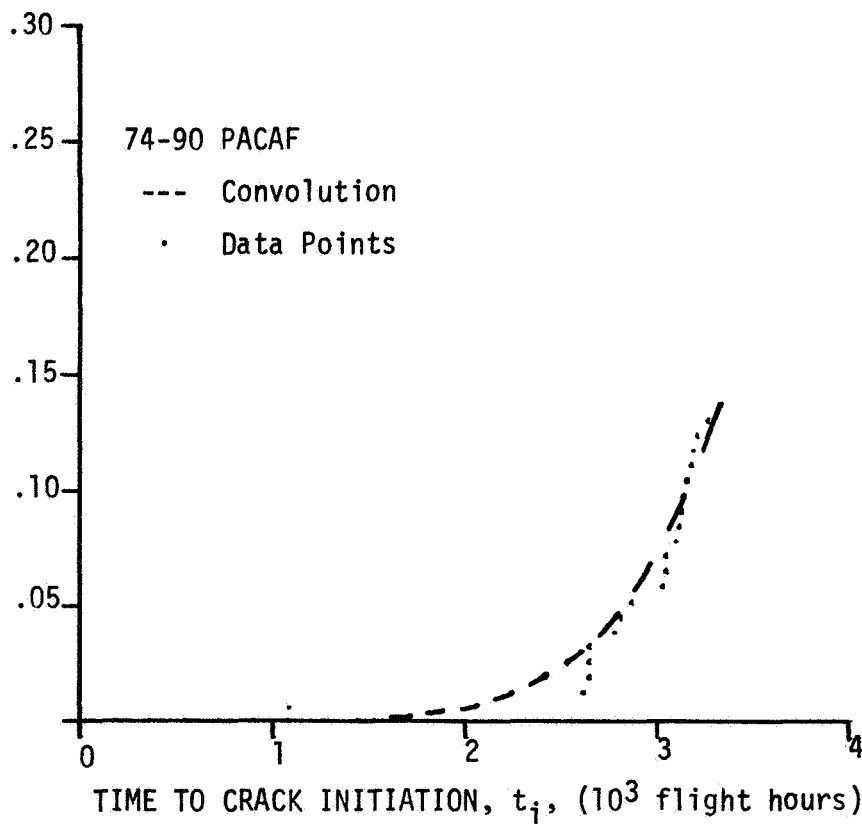


Fig. 34. Lower Tail of Crack Initiation Time Distribution.

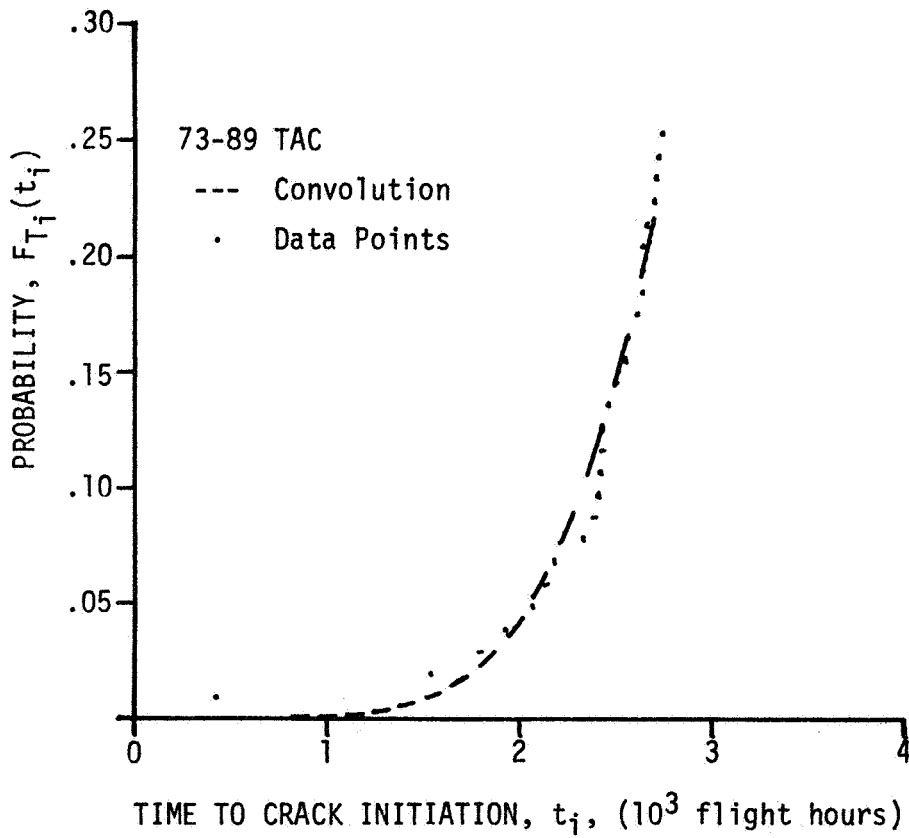


Fig. 35. Lower Tail of Crack Initiation Time Distribution.

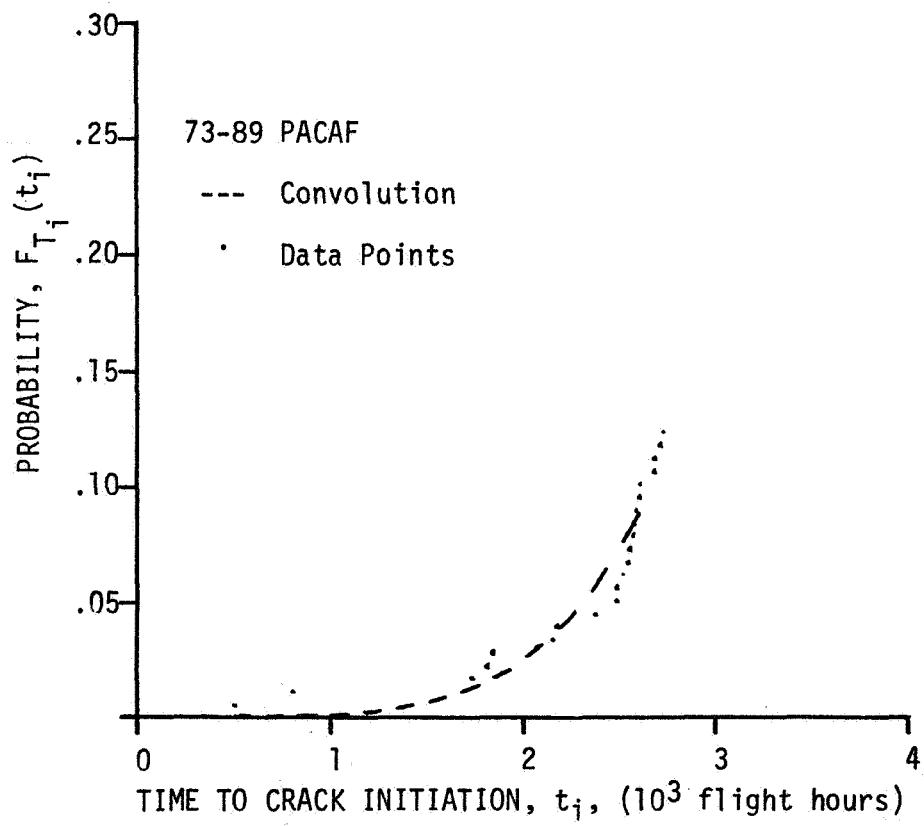


Fig. 36. Lower Tail of Crack Initiation Time Distribution.

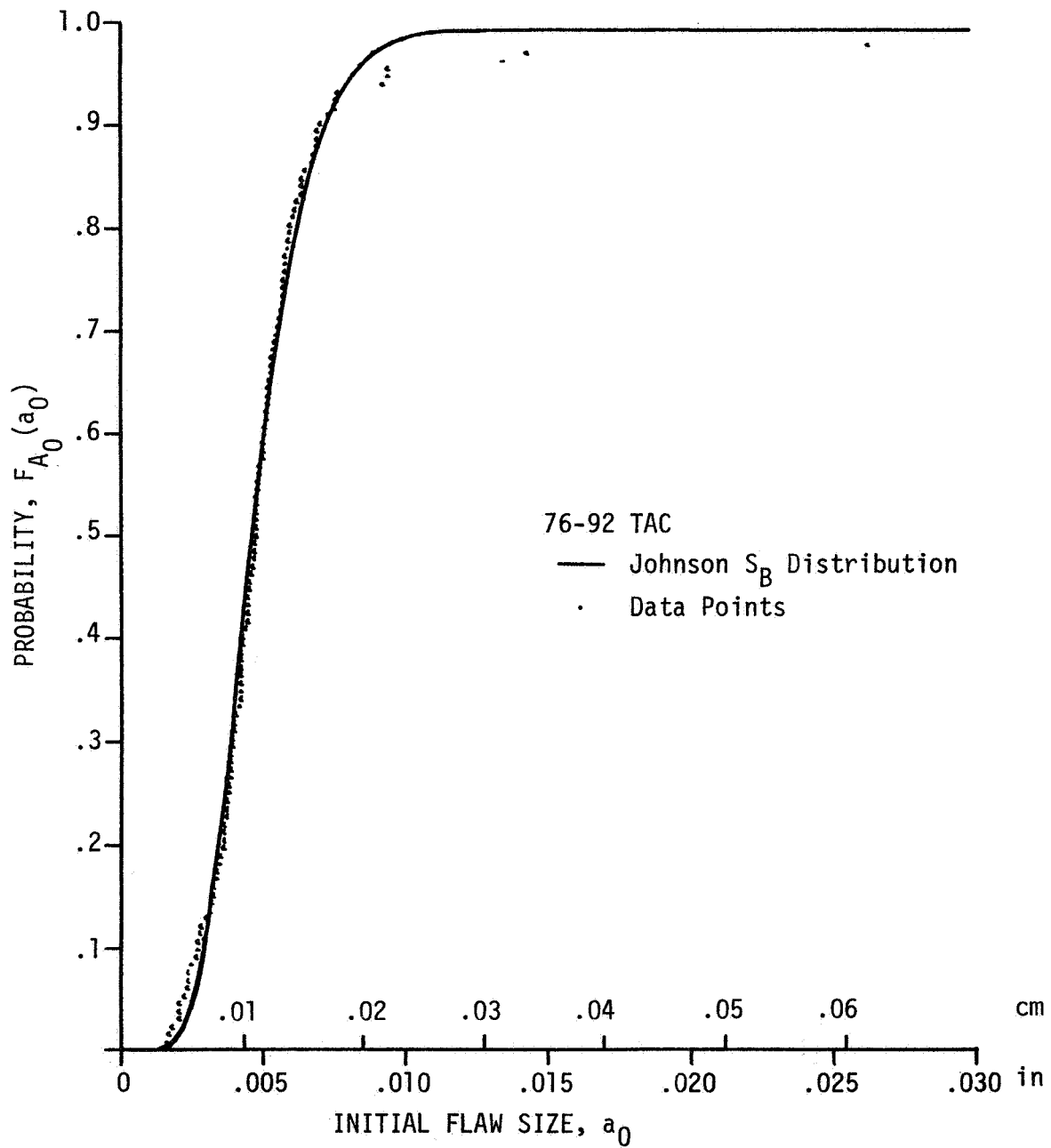


Fig. 37. Distribution of Initial Flaw Sizes.

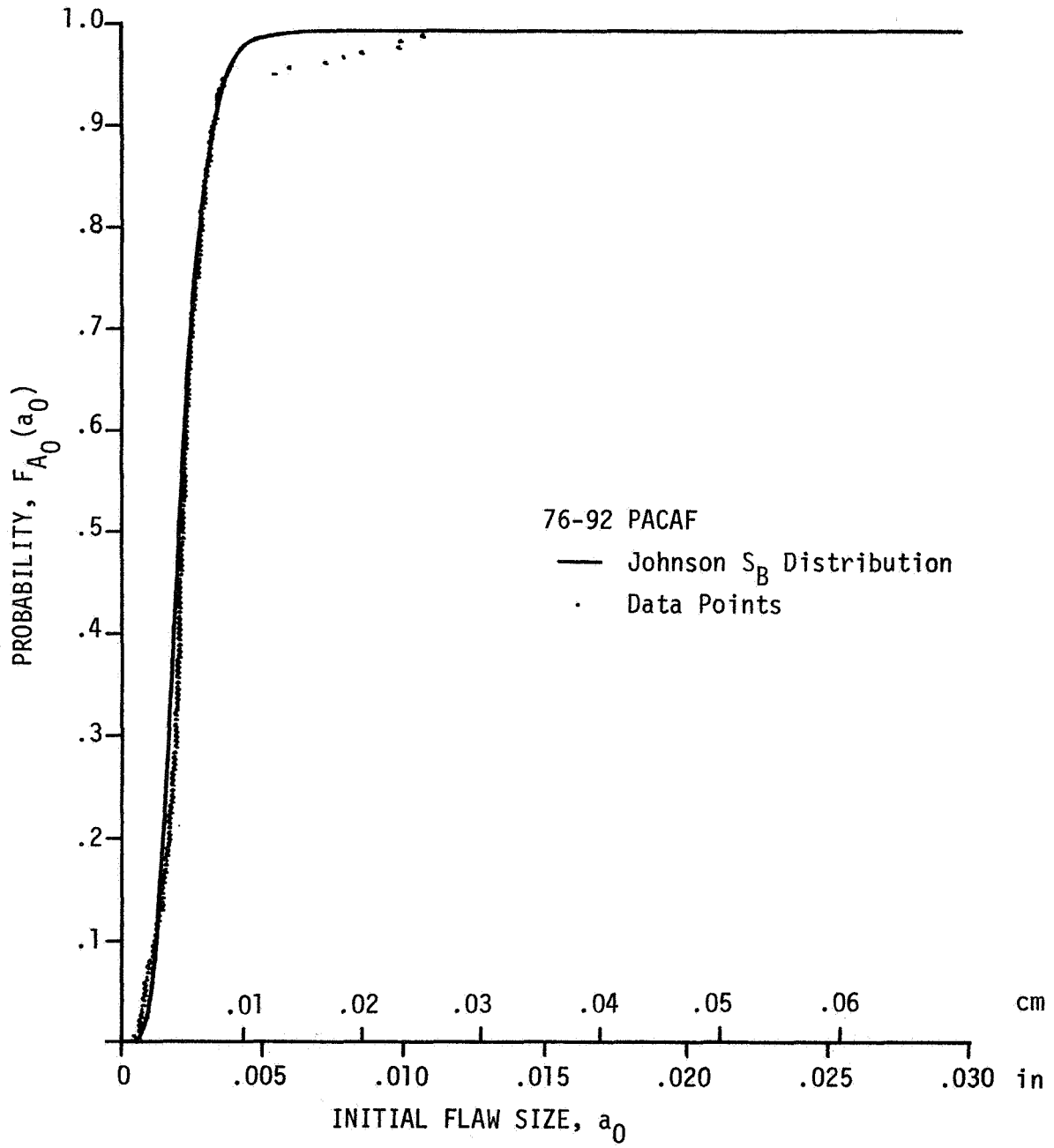


Fig. 38. Distribution of Initial Flaw Sizes .

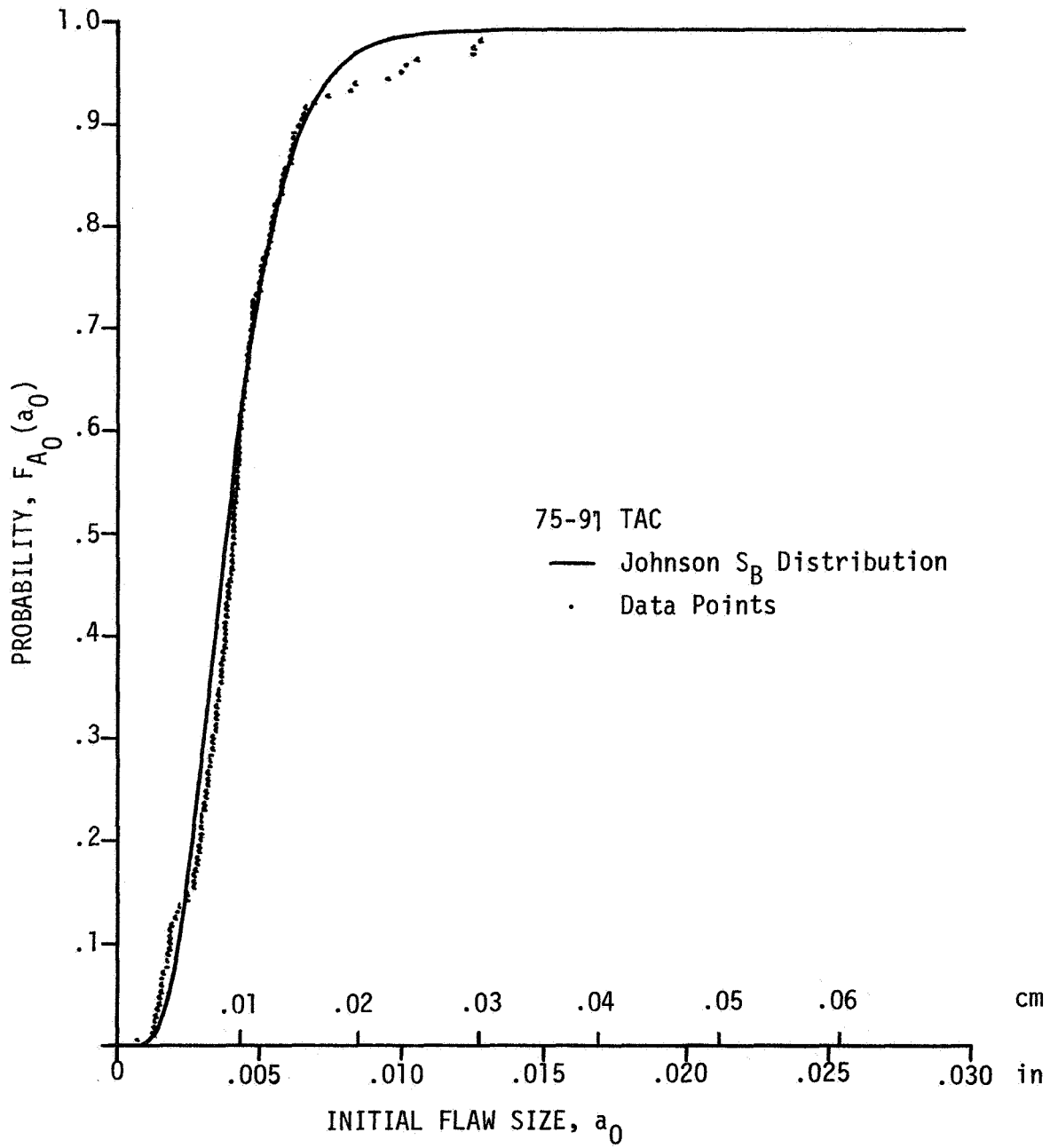


Fig. 39. Distribution of Initial Flaw Sizes.

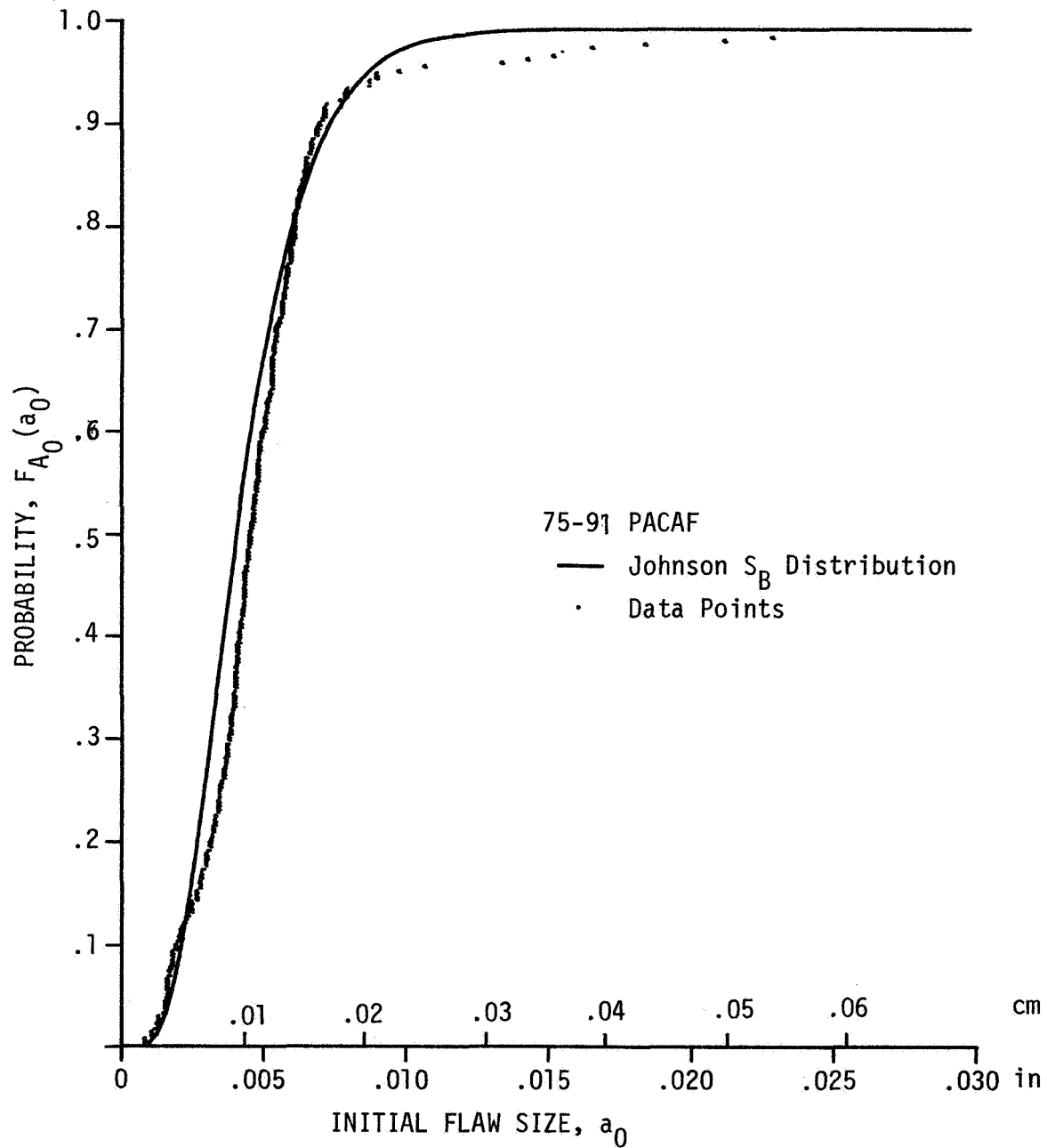


Fig. 40. Distribution of Initial Flaw Sizes.

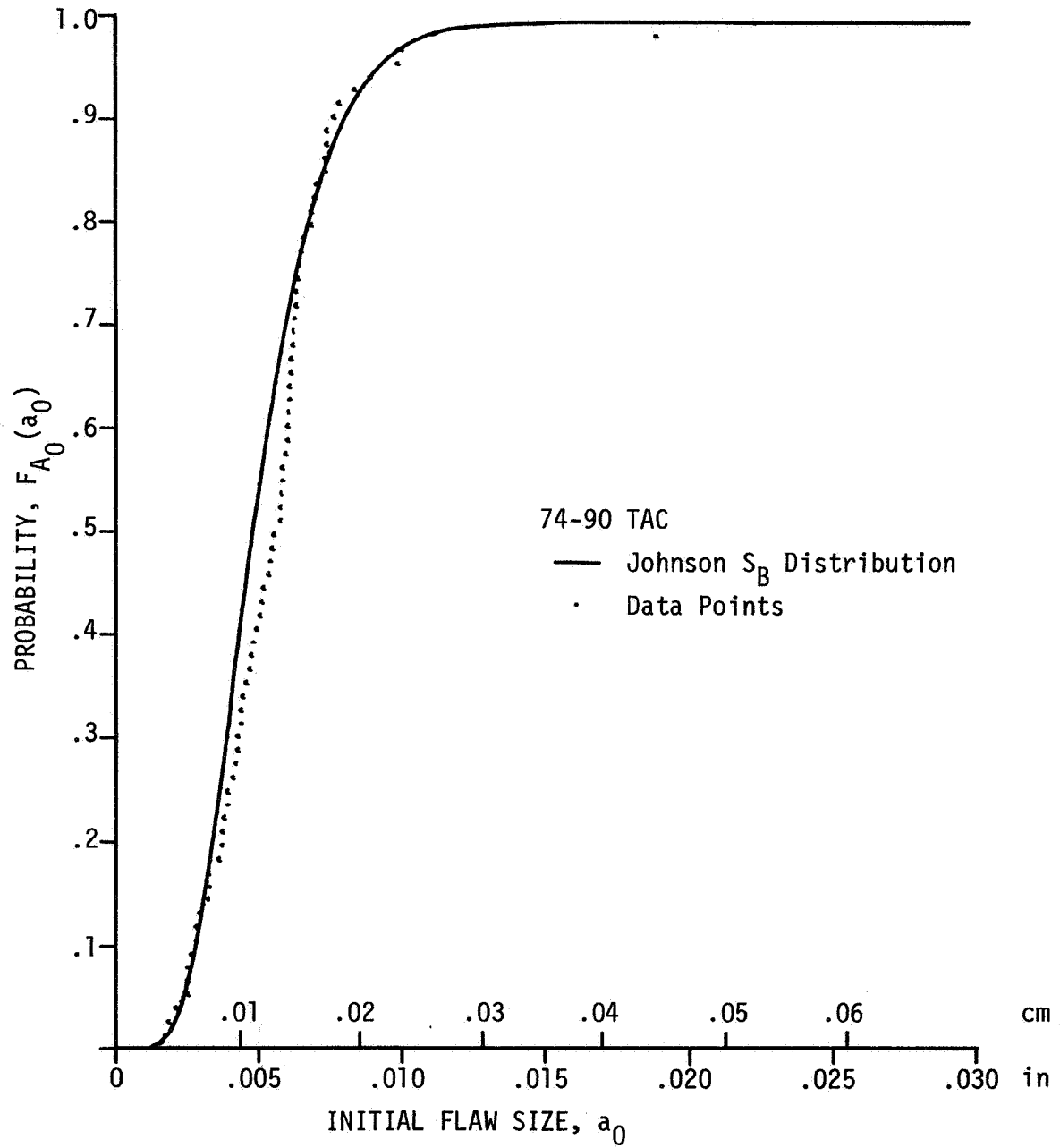


Fig. 41. Distribution of Initial Flaw Sizes.

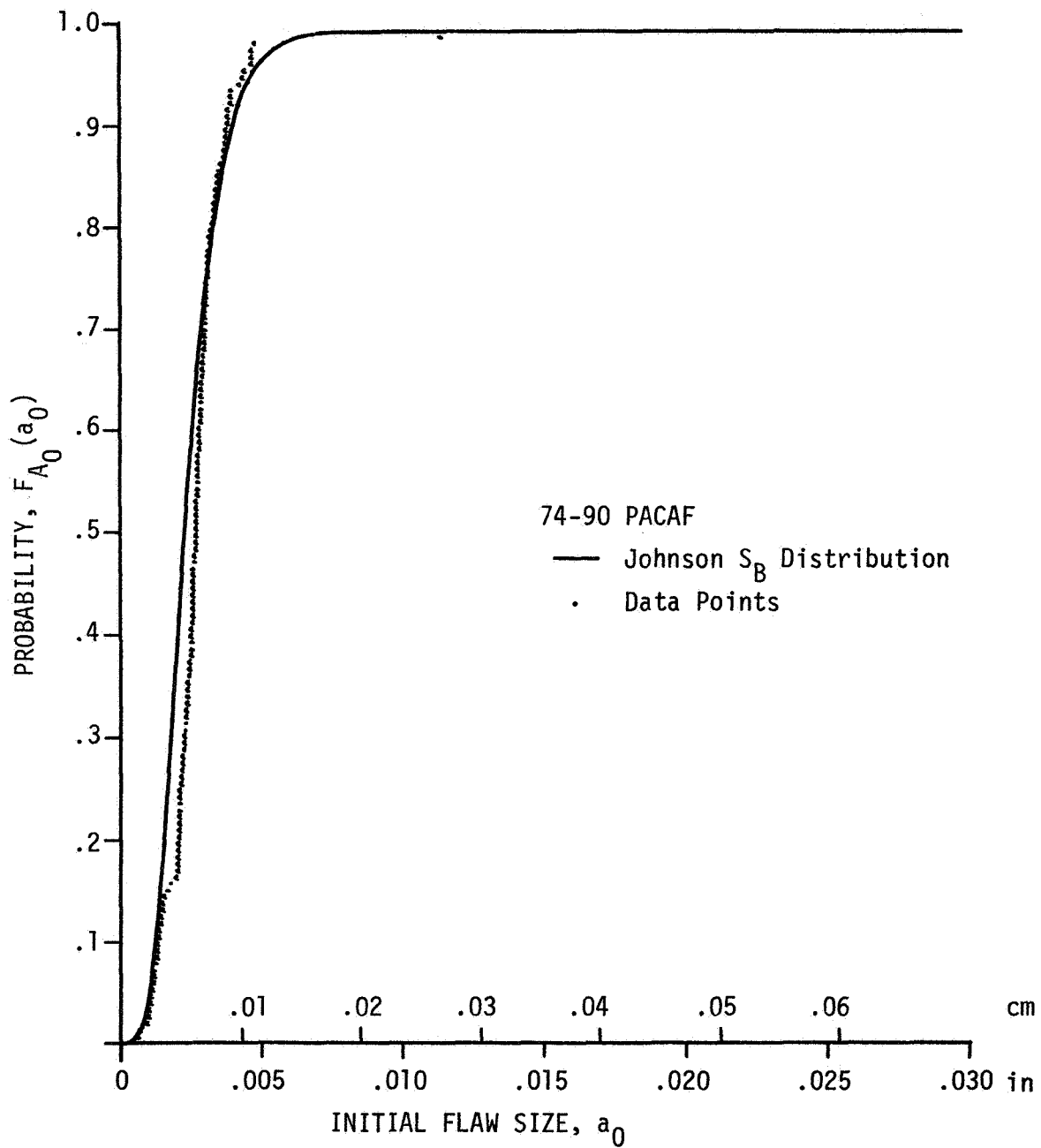


Fig. 42. Distribution of Initial Flaw Sizes.

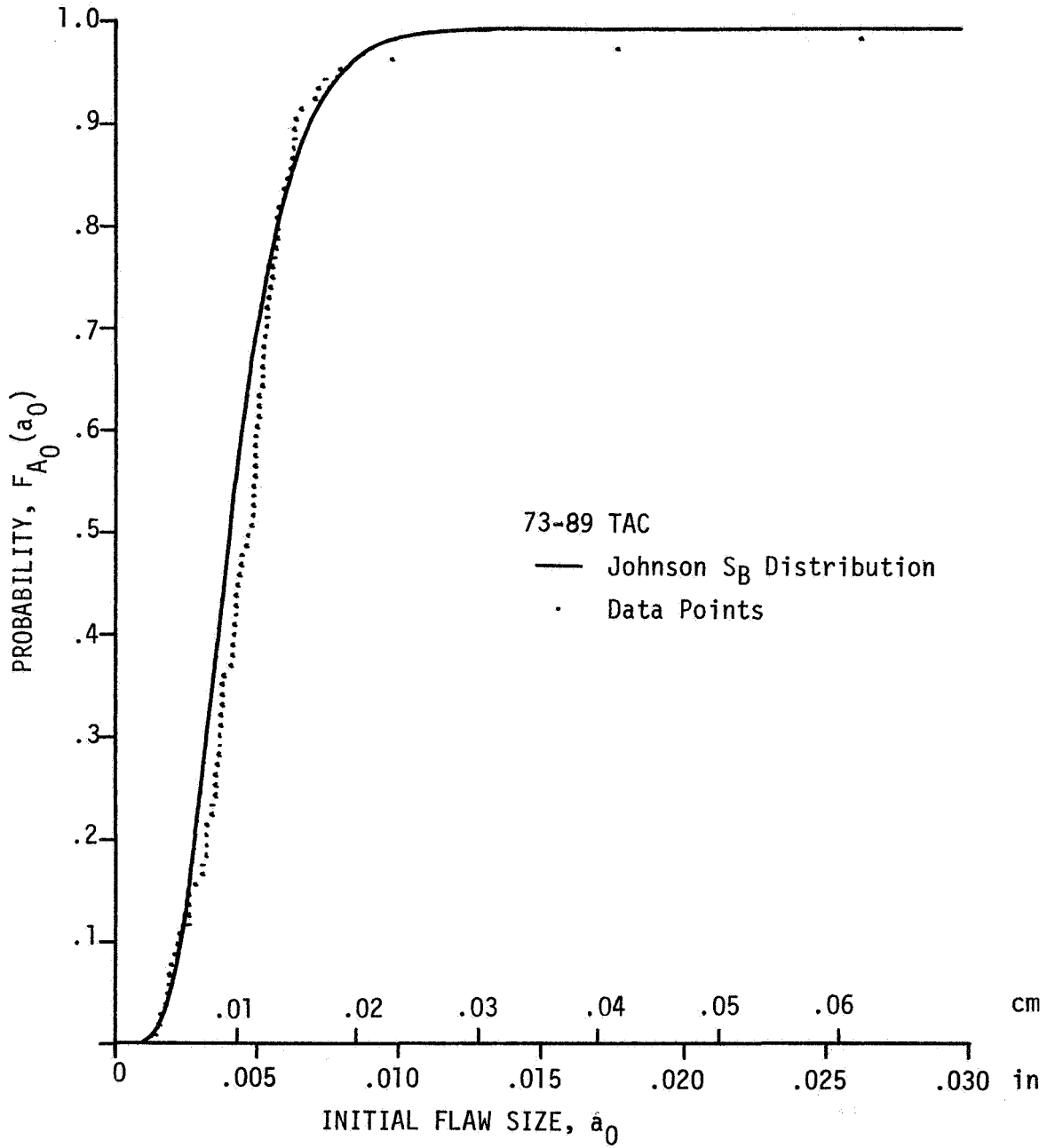


Fig. 43. Distribution of Initial Flaw Sizes.

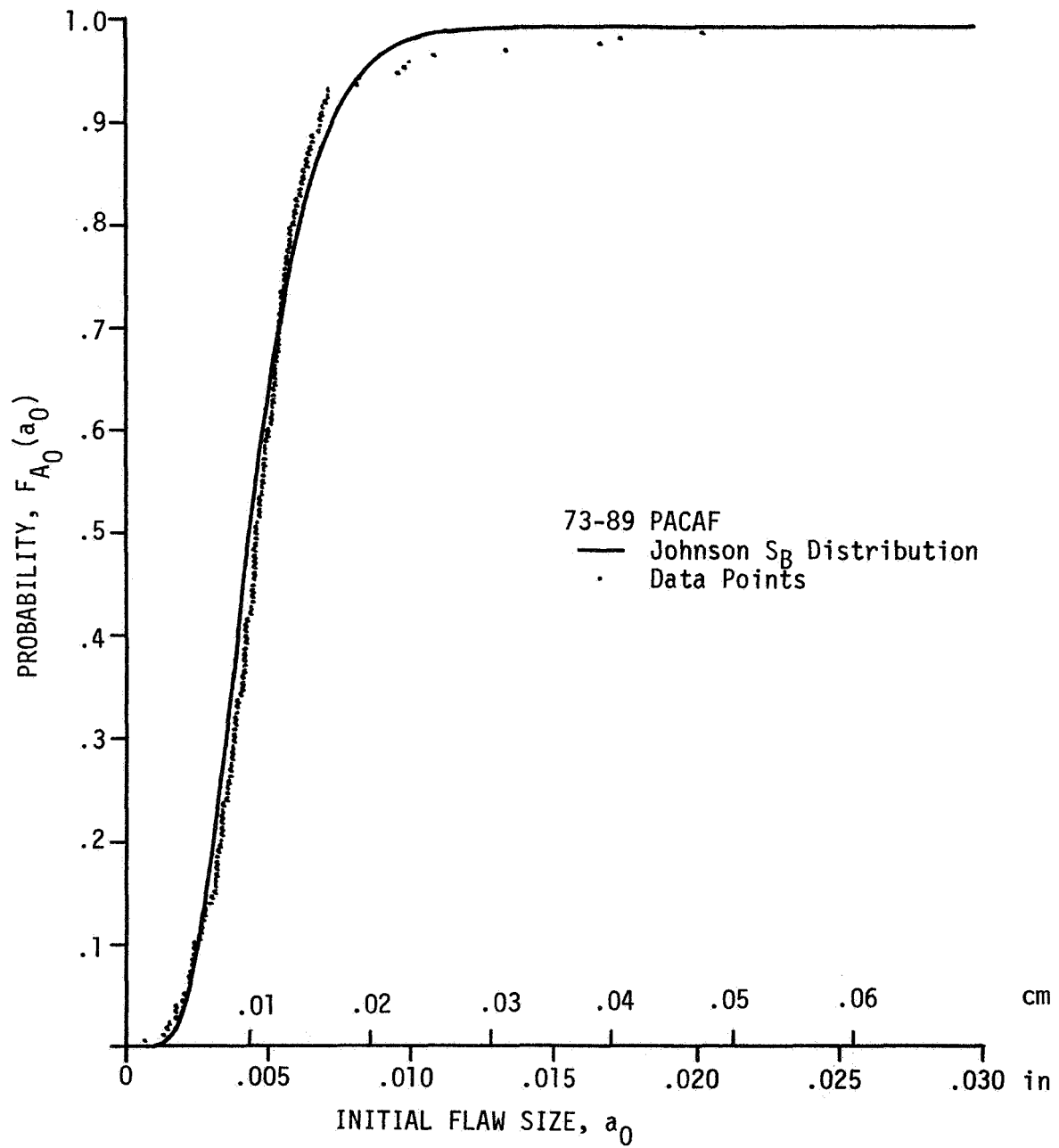


Fig. 44. Distribution of Initial Flaw Sizes.

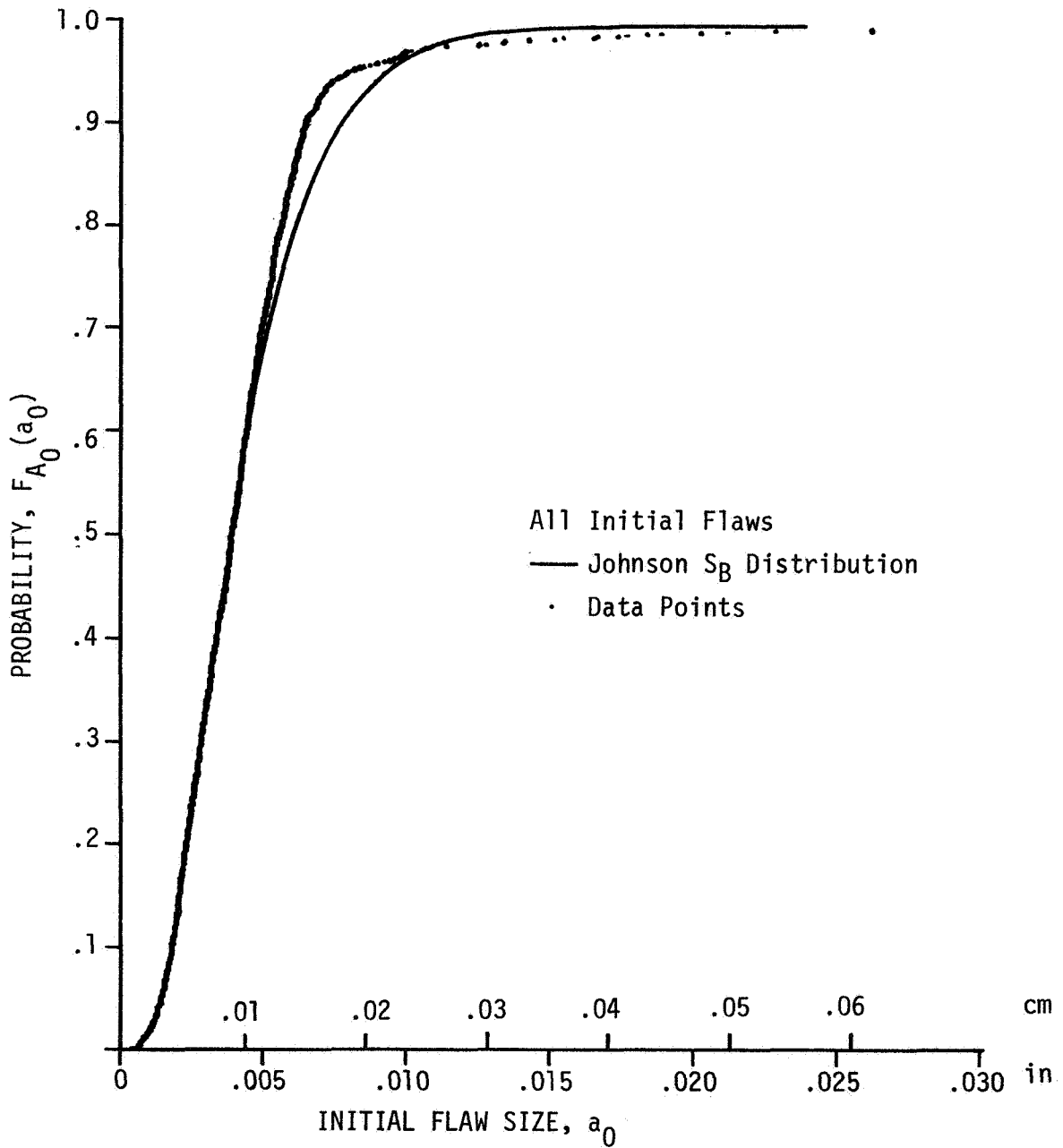


Fig. 45. Distribution of Initial Flaw Sizes for all 1256 Growing Cracks.

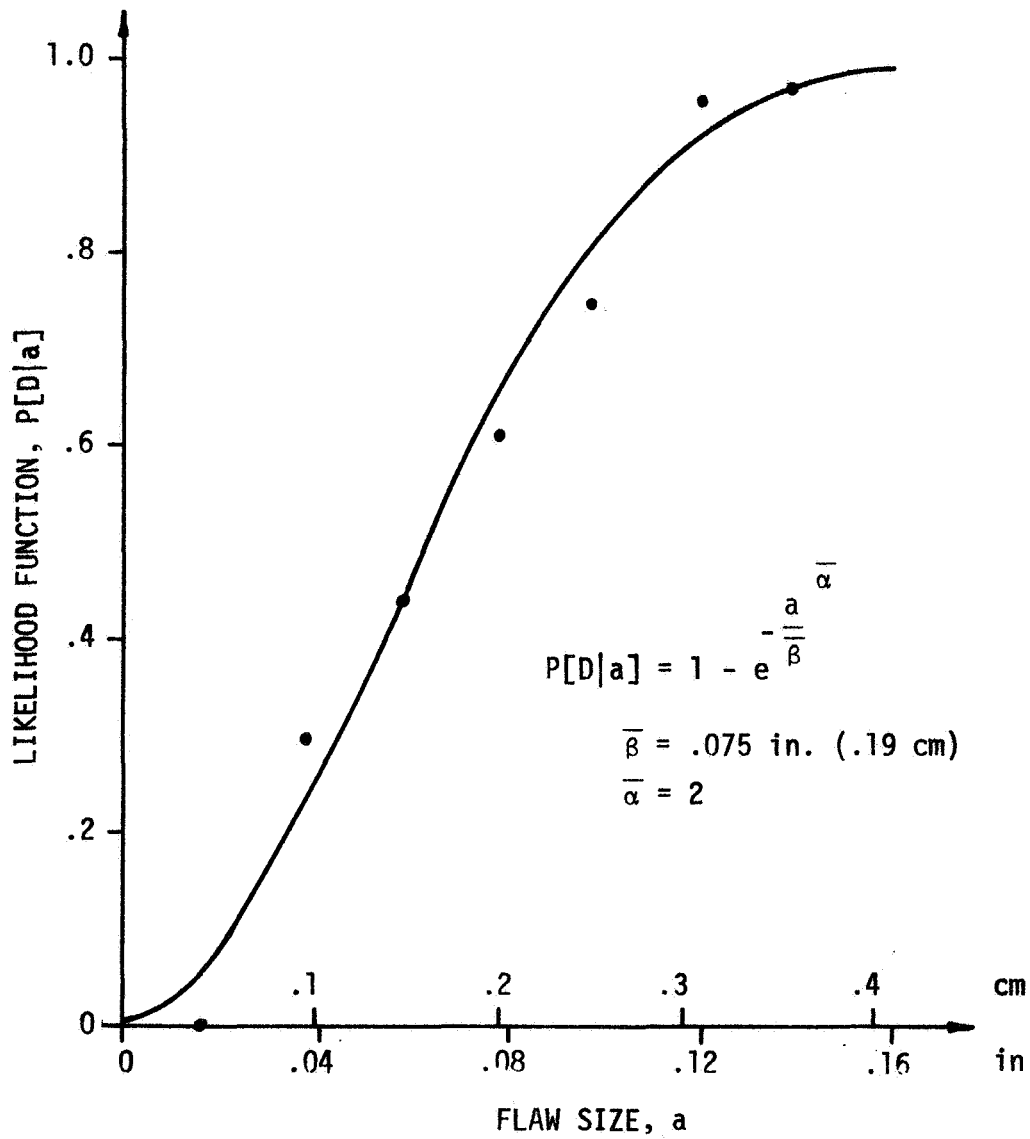


Fig. 46. Probability of Detection, "Likelihood Function".

Data From Ref. 26.

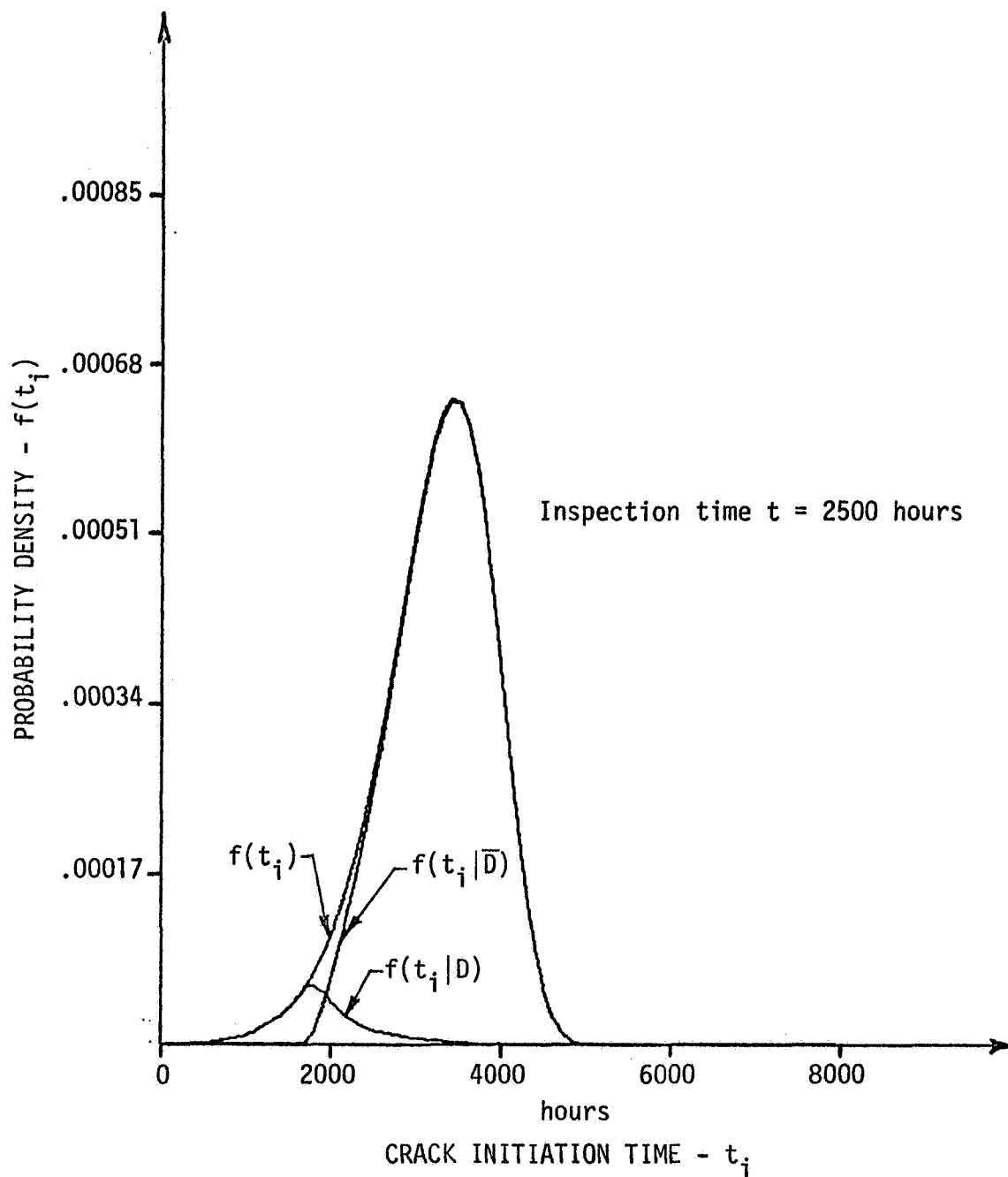


Fig. 47. Distributions of Crack Initiation Times for a Typical Wing Location for an Early Inspection.

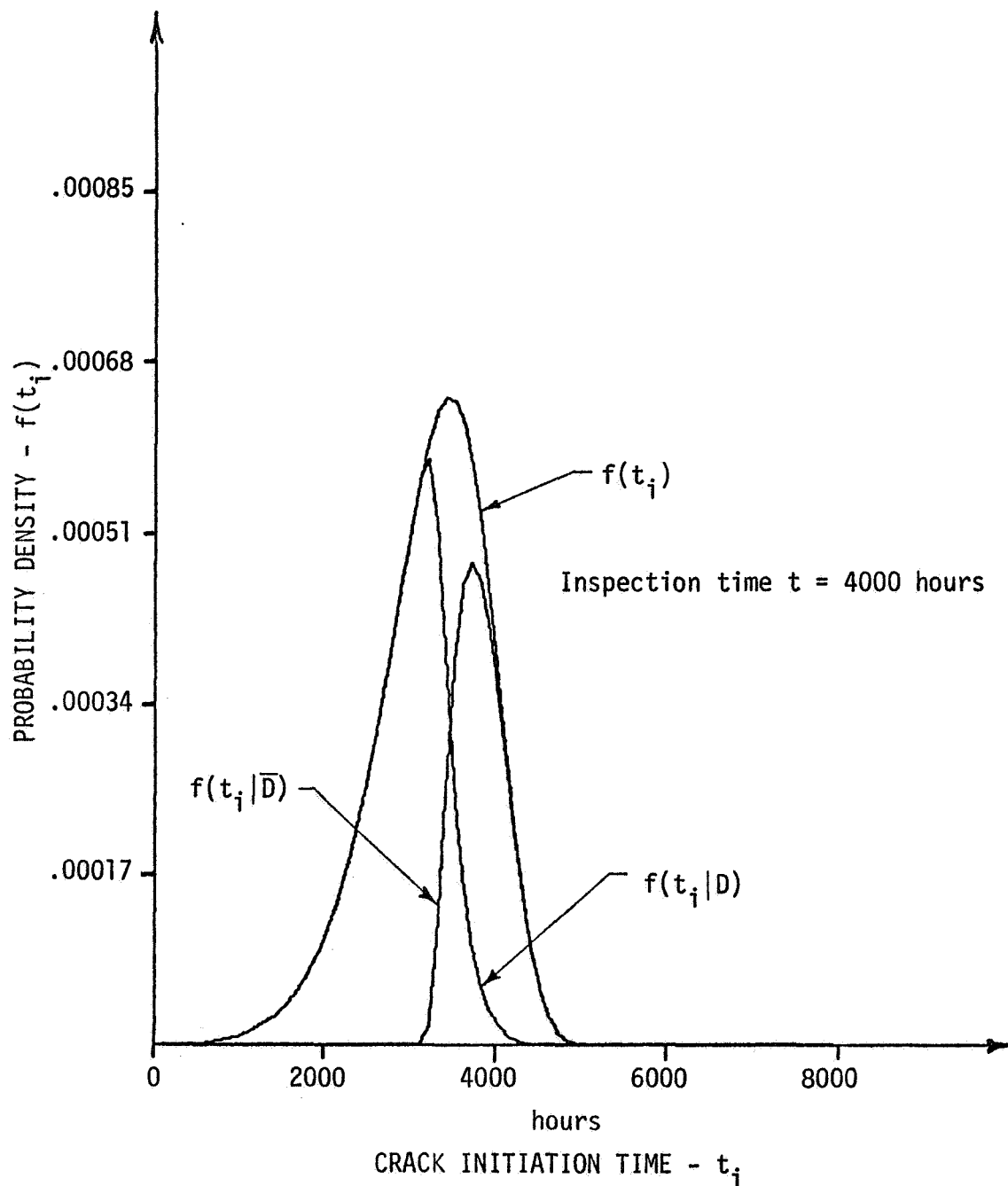


Fig. 48. Distributions of Crack Initiation Times for a Typical Wing Location for an Intermediate Inspection.

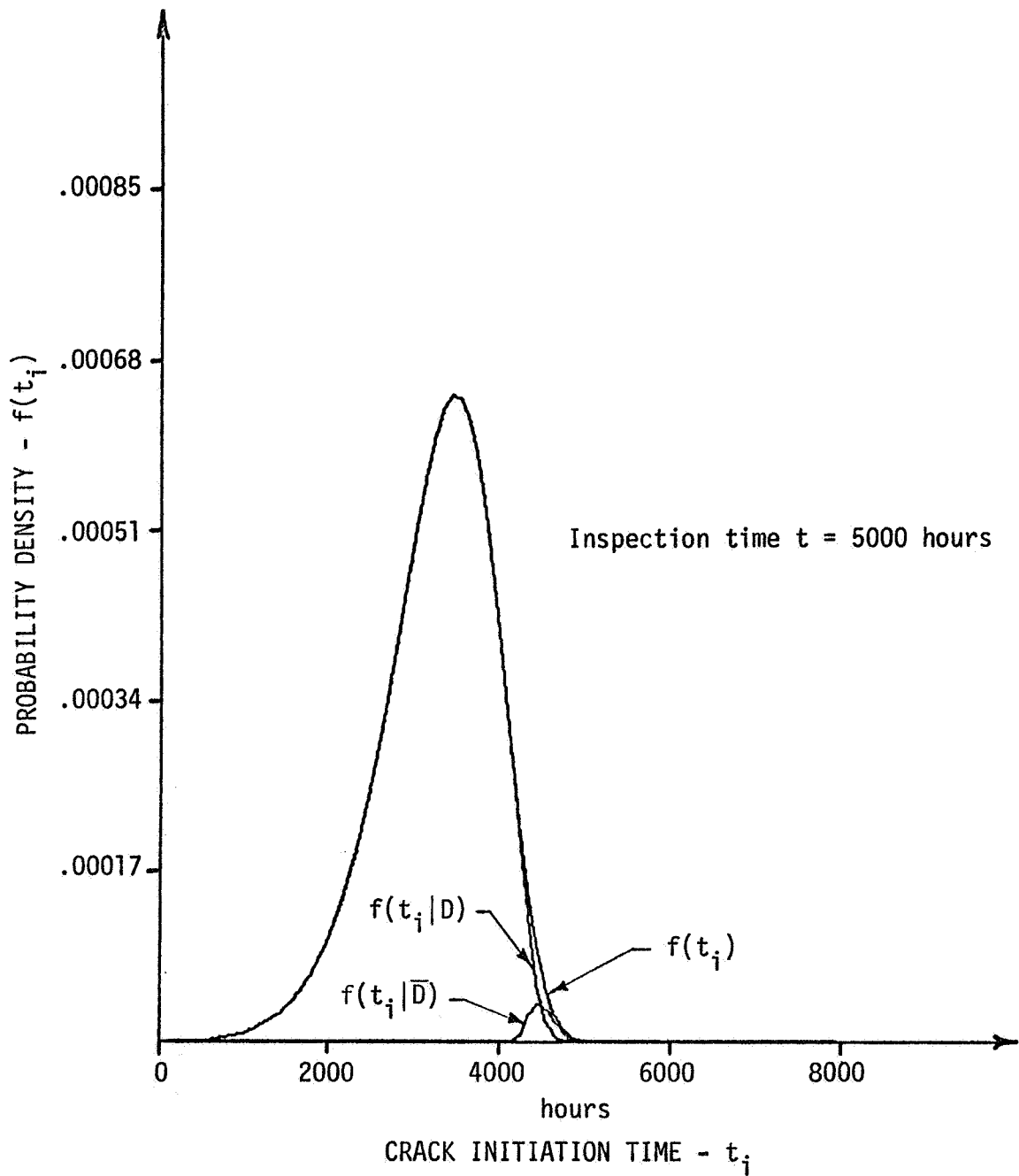


Fig. 49. Distributions of Crack Initiation Times for a Typical Wing Location for a Late Inspection.