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**ANTENNA EVALUATION STUDY  
FOR THE  
SHUTTLE MULTISPECTRAL RADAR:  
PHASE I**

*CR 151232*

**FINAL REPORT**

by

**Edgar L. Coffey, III  
Keith R. Carver**

prepared for

**NASA Johnson Space Center  
Houston, Texas**

**Contract No.  
NAS 9-95469**

**December 1976**



**Physical Science Laboratory**

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(NASA-CR-151232) ANTENNA EVALUATION STUDY  
FOR THE SHUTTLE MULTISPECTRAL RADAR, PHASE I  
FINAL REPORT (NEW MEXICO STATE UNIV.) 204 P  
NO 210/52 VOL 1 501 171

M77-19269

UNCLAS

G3/52 20008

PA00874

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## 1.0 INTRODUCTION

This report presents the results of the first phase of the Antenna Evaluation Study for the Shuttle Multispectral Radar (SMR). The original goals of Phase I were to make preliminary identification of those critical parameters of the Shuttle Multispectral Radar Antenna (SMRA) which most affect antenna performance and to write a first draft of specifications for the subarrays which will be analyzed in Phases III and IV. During the course of this research it was decided to expand the scope of Phase I to include the development of a preliminary mathematical model describing SMRA performance under the influence of various physical and environmental factors. This was necessary to identify those key antenna parameters and environmental conditions which might degrade SMRA performance. This has allowed more time in Phase II for the simulation and study of critical factors which may create or influence these error-causing conditions.

### 1.1 Some Projected User Needs

A key and as yet not completely answered question is the instrumentation requirement of the rather nebulously defined user community. Studies by Hughes Aircraft Co. and JPL have suggested L and X-bands as a dual-frequency choice best suited to the needs of potential users, yet realizable in terms of development. In addition, dual polarization capability has been specified along with three incidence angles and various swath widths.

Space radars with imaging capabilities have a considerable potential applicability in the following areas:

1. Soil moisture determination
2. Flood area monitoring
3. Crop discrimination
4. Crop yield estimates
5. Plant biomass
6. Mineral and petroleum exploration
7. Sea and lake ice mapping
8. Iceberg monitoring

These application areas have been discussed in two Active Microwave Workshops (Fall, 1974 and Fall, 1976 at NASA/JSC) along with an additional October, 1976 Shuttle Multispectral Radar Applications Meeting held at NASA/JSC. These meetings have also pointed out that L and X-bands are not optimum for some prime application areas. For example, C-band and  $K_u$ -band are optimum for soil moisture measurement and vegetation classification, respectively. At the time of this writing, the user groups have not been able to agree on two optimum frequencies best suited for the broadest range of application; this has imposed additional delays on detailed specification of optimum incidence angles, registration of cross-band beams, etc. For example, it has been shown that soil moisture determination is best carried out at about 4.5 GHz and for incidence angles in the  $7^\circ - 12^\circ$  range, whereas crop classification is optimally accomplished at 14 GHz and with a  $40^\circ - 50^\circ$  incidence angle range.

This situation of uncertainty requires that the present antenna study incorporate frequencies ranging from 1.2 - 14.5 GHz as well as a consideration of incidence angles from near-nadir to nearly  $50^\circ$ . In fact, one of the eventual goals of this project is to provide simulated performance data at a variety of frequencies, incidence angles, etc.; this may impose additional hardware-related restrictions on finalized choices of SMR design parameters.

### 1.2 The Need For A Comprehensive and Independent Antenna Evaluation Study

The Shuttle Multispectral Radar Antenna, being an electrically large radiator, has several characteristics that require a thorough analysis. Preliminary studies indicate that the SMRA must be a large structure, on the order of 12 m by 3 m which can operate in a space environment. The determination of its electrical behavior (gain, beamshape, pointing accuracy, etc.) is vital to the calibration of the overall SAR system and thus to the success of the mission itself. Since the antenna development cost is by far the largest system development budget item, it is economically imperative that critical antenna parameters and potential calibration problems be identified early in order to avoid costly mechanical and electrical redesign phases



which might otherwise be required. In effect, a macro developmental and simulation exercise such as described herein seeks to clearly identify at a very early stage those most likely technological problem areas which invariably accompany the development of a state-of-the-art instrument such as the Shuttle Multispectral Radar; this not only helps to avoid costly development errors but also permits a realistic prediction of the performance of the instrument itself and thus allows design/performance tradeoffs to be made at a much earlier stage of development.

The huge electrical size of the total antenna structure suggests that conventional techniques for measurement of important electrical parameters will be of little use and that novel calibration schemes must be devised. In addition, electrical and mechanical tolerance errors, particularly systematic ones, are more likely to seriously degrade the performance of a large antenna. In preliminary investigations by Hughes Aircraft Co. and Ball Brothers Research Corp., the antenna configuration and size was specified, but no bounds were placed on tolerance errors or performance indices. These bounds should be determined as soon as possible to avoid the expense of constructing either an antenna whose parameters are overly restricted or an antenna whose performance will be inadequate for the desired applications.

In addition to determining antenna performance in an idealized environment, it is also necessary to predict in situ performance and to devise means for assurance of calibration stability. This prediction requires an adequate mathematical model which describes the SMRA performance including space environmental factors such as thermal gradient effects on electrical flatness, beam pointing stability, material stability in a near-vacuum environment, and effects of multiple scatter from the shuttle bus itself.

Although the effect of the SAR processor on the synthetic antenna performance has not been considered in this report, it will be discussed in subsequent reports. The quality of the image processors currently being considered may have little effect on the image if the antenna performance is significantly degraded by external factors, or vice versa. A related consideration is the study of realistic system calibration techniques. System development must

be planned with calibration as an integral part of the design process; calibration must not be an afterthought. Since the SMR can only be calibrated to within a specified uncertainty, the antenna should not be overspecified to a level which cannot even be measured. For example, it should be possible to answer the question, "What level of antenna cross-polarization rejection is actually necessary in light of: 1) user requirements, and 2) ability to calibrate cross-polarized performance?"

### 1.3 A Two-Path Approach To The Antenna Study Problem

An answer to the questions in the last section requires the development of both mathematical techniques and computer programs to analyze random and systematic electrical and mechanical SMRA errors, and a method of measuring and testing antennas representing the competing approaches. This dual approach has been followed in Phase I and will continue to be used in subsequent phases.

The mechanical model and computer simulation relate electrical and mechanical tolerances to performance indices. The model is being designed so that tradeoffs can be made between the quality of performance and the expense of tight tolerances. It is also capable of predicting SMRA performance under various space environmental conditions. Finally, through analysis of the results of several simulations, the model is able to flag any possible problems that one of the competing designs might incur. In Phase I antenna parameters have been identified which are critical to both antenna and system performance. Furthermore, a preliminary mathematical model and a computer program have been developed and tested. Subsequent phases will deal with refinement of the model and the simulation of a number of scenarios.

The measurement/testing problems can be alleviated by performing appropriate tests on macro model subarray antennas representing the competing design approaches. These measurements will include near-field and far-field electrical performance tests and thermal cycling tests with mechanical flatness measurements. The results of these tests will be entered into the SMRA simulation model to predict the performance of the full size array. In Phase I, those parameters which must be measured have been identified.

#### 1.4 Scope of Work

This report presents the results of several activities associated with the development of the mathematical model, the computer program, and the subarray panel requirement. These activities were:

1. A review of the antenna designs from the definition phase.
2. An identification of critical SMRA performance parameters.
3. An identification of conditions which may degrade SMRA performance.
4. An identification of factors which may create or influence error-causing conditions.
5. The development of a preliminary SMRA mathematical model.
6. The organization of the SMRA computer simulation program.
7. A demonstration of the flexibility of the SMRA computer program.
8. The identification of requirements and specifications for the subarray panels.
9. The development of an interface data tape for the University of Texas Applied Research Laboratory image processor simulation program.

The principal results obtained to date are:

1. A cause-effect relationship for various environmental and physical factors on the antenna electrical and mechanical behavior.
2. A baseline mathematical model for the electrical and mechanical behavior of the SMRA in a space environment.
3. A compilation of results of the simulation of the baseline designs for both competing approaches and several sample scenarios demonstrating the flexibility of the mathematical simulation.

## 2.0 IDENTIFICATION OF IMPORTANT PARAMETERS, THEIR EFFECT ON SMR SYSTEM PERFORMANCE AND POSSIBLE TRADEOFFS

### 2.1 A Review Of The Antenna Designs From The Definition Phase

A review of the candidate SMR system approaches and synthetic aperture techniques was conducted early in Phase I in order to establish a familiarity with both the antennas themselves and the interaction between the antenna and the image processor. It was found that although preliminary specifications had been placed on some of the antenna parameters, there was not a tolerance budget (or rationale for one) for each major component of the overall system. Because the performance of the overall system is set by both the antenna and the image processor, and because these cannot be designed independently, it is our opinion that further refinement of antenna performance parameters be withheld pending dual simulation of antenna and processor. To this end, we have developed an auxiliary computer program which generates real antenna pattern data for use by the Applied Research Laboratories (University of Texas) in their image processor simulations.

Both Hughes and JPL have suggested in independently conducted studies that the SMR antenna be a large, planar array of approximately three meters by twelve meters. The twelve meter azimuth dimension was predicated upon the resolution requirement. The three meter elevation dimension was based upon the swath width requirement, space required for both L- and X-band antennas, and payload weight/size restrictions. Both designs would use uniform illumination and would achieve three different elevation beamwidths by switching between two panels. The angle of incidence would be controlled by the attitudes of the shuttle. Hughes proposed to use crossed dipoles at L-band and waveguide slots at X-band, whereas Ball Brothers (under contract to JPL) would use microstrip elements at both frequencies. For launch and boost, the antenna would be folded twice in the azimuth dimension and stowed within the shuttle payload compartment. After orbit insertion, the antenna would be deployed to its nominally flat shape. A comparison of the Hughes and JPL/Ball Brothers antenna specifications is given in Table 1. More detailed information can be found in the Hughes report to NASA/JSC of October 1975 and the JPL report to NASA/JSC of March 1976.

TABLE 1. A comparison of JPL/Ball Brothers and Hughes antenna specifications.

Specification	JPL/BB			HUGHES		
	L-band	X-band		L-band	X-band	
Frequency, GHz	1.302	8.33		1.04	9.00	
Polarization	HH, HV, VH, VV	HH, HV, VH, VV		HH, HV, VH, VV	HH, HV, VH, VV	
Azimuth dimension, m	12	12		12	12	
Azimuth Peak-to-Null beamwidth, deg	1.1	0.172		1.287	0.153	
Efficiency, %	70	70		---	---	
Peak power rating, KW	6	20		6.8	17	
Pointing accuracy, deg	+0.5	+ 0.5		---	---	
Peak sidelobe level, dB	13.5	13.5		13	13	
Element type	microstrip	microstrip		crossed dipoles	waveguide slots	
Off-nadir angle, deg	25 38 50	25 38 50		10 - 60	10 - 60	
Peak gain, dB.	32 36 38	41 45 46		29* 32* 34*	38* 42* 43*	
Elevation dimension, m <sup>†</sup>	0.65 1.55 2.2	0.12 0.24 0.36		0.81 1.62 2.53	0.094 0.188 0.282	
Elevation Peak-to-Null beamwidth, deg.	20.8 8.5 6.0	17.45 8.63 5.74		18.87 9.44 6.29	18.87 9.44 6.29	

\*Gain figures include 2.4 dB system loss

†The two panels are combined to form the narrow beam panel

The similarity of the two designs makes it possible to model both antennas with the same algorithm and computer program. The only significant electrical difference is the element type, and this may be handled with only a minor change in one subprogram.

## 2.2 Critical Antenna Parameters

Six antenna electrical parameters have been identified as being critical to the SMR performance. They are:

1. Antenna gain and efficiency.
2. Main beam shape.
3. Side lobe level.
4. Polarization purity.
5. Beam pointing accuracy.
6. Cross-band and cross-polarization beam coincidence.

For each of these parameters, both static errors and dynamic errors must be considered. In many cases, static errors, or those errors which do not change during a mission, can be compensated. On the other hand, dynamic errors whose effect on performance may change during a mission, are quite difficult to handle. For example, the shape of the main beam will modulate the data. If the beam shape is known, this effect may be processed out of the final image. But if the beam shape changes during flight, as it may due to thermal distortion of the antenna structure, it may be impossible to compensate for the time-varying modulation without real-time knowledge of the antenna pattern.

In the following, each of the above six parameters will be discussed considering both types of errors and possible tradeoffs with other system components.

Antenna Gain & Efficiency -- Static errors will affect the system signal-to-noise ratio (SNR). Tradeoffs are with transmitter power, receiver sensitivity, and minimum detectable scattering cross section. Slow changes in the antenna gain or efficiency over a period of hours or days cause instability in the system calibration.

Main Beam Shape -- The azimuth beam shape does not affect the system as much as the elevation beam shape, since each resolution cell "sees" the entire azimuth beam but only sees a portion of the elevation beam. As mentioned above, the static antenna gain pattern weights the amplitude of the data, but this effect can be compensated by using the appropriate (in elevation) processor demodulation algorithm. However, dynamic changes in the beam (e.g., caused by thermal deformations related to changing sun angle) are largely uncorrectable and lead to a degradation in image quality.

Side Lobe Level (SLL) -- The principal impact of the side lobe level on the overall system is one of ambiguities within the processed image. Uniform illumination of the SMRA will produce a theoretical SLL of -13.3 dB (one-way), but this level may rise due to panel folding/unfolding errors, mechanical and thermal distortions, etc. However, azimuth ambiguities can be suppressed further by proper choice of the Doppler processing bandwidth and the pulse repetition frequency (PRF). Range ambiguities are the result of ground point returns that are outside the desired image swath width which arrives at the receiver simultaneously with the return from a point which lies within the desired swath width. Unfortunately, their control is provided almost exclusively by the antenna elevation pattern. Consequently, any rise in the side lobes, static or time-varying, will decrease the imaged signal-to-ambiguity ratio. Furthermore, an increase in the SLL indicates that the antenna gain has decreased, degrading the SNR as well. Elevation side lobes are the most critical since no compensation can be performed on the processed image.

Another quality criterion is that of total peak to total side lobe power ratio. Even though all side lobes may fall below some relative level (for example, -20 dB), the integrated SLL power level may completely mask the presence of a fairly strong point target.

One obvious, but expensive, solution to the ambiguities problem would be to overdesign the range pattern of the SMRA by tapering the range excitation amplitude to produce an even lower side lobe level. There are two other tradeoffs to be considered. First, the SMRA would be more susceptible to errors induced by mechanical and thermal distortions since tighter control on

magnitude and phase excitation tolerances is necessary to obtain the low side lobe pattern. Secondly, the three required beamwidths could not be obtained simply by switching between the two subarrays as in the case of uniform illumination.

Polarization Purity -- No adequately substantiated performance guideline for polarization purity has yet been announced. However, it is felt that this parameter will be important to the scientific community and should be considered. In theory, the presence of any cross-polarized field at the antenna output will degrade the quality of the data. However, the cross-polarization level performance actually required depends on the intended use of the data and the sophistication of the  $\sigma^0$  classification models. For example, in the X-band GEMS SAR, most users feel that a -20 dB cross-polarization level is quite adequate. The level of this unwanted component can be controlled to some degree by the selection of the array element type and the fabrication of a supporting structure that will allow little, if any, deviation of the array surface from mechanical flatness. The principal tradeoffs are against expense and technical risk. It is conceivable that any static purity problems might be compensated in the processing hardware, but certainly, time-changing instabilities would be uncontrollable.

Beam Pointing Accuracy -- The items basic to the subject of antenna pointing are: 1) The position of the antenna beam relative to the zero Doppler plane; and 2) The rate of change of that position. Both of these items must be considered when attempting to locate the plane of zero Doppler for the processor. In the process of forming the synthetic aperture, it is necessary to sense any changes in the position of the antenna beam relative to the isodops (surfaces of constant Doppler frequency) so that the resulting image may be compensated. One way of doing this is to monitor in real time the average Doppler shift of the radar data and use this information to keep the beam centered about the required isodop. Unfortunately, this requires precise control of the beam pointing direction, and any errors, static or dynamic, will tend to invalidate this approach.

A second approach would be to monitor the data and dynamically adjust the processor to compensate for deviations in beam position as well as orbit



eccentricity and angular velocity (combined pitch, roll, and yaw velocities) of the spacecraft. Using this approach, the antenna requirements are reduced (at the expense of increased processor complexity) to placing limits on the angular excursions of the combined antenna electrical/mechanical beam pointing directions and on the rate of change of this angular motion.

The following antenna pointing and stability limits have been recommended<sup>†</sup>:

± 0.5 degrees in pitch, roll and yaw.

0.01 degrees per second maximum in pitch, roll, and yaw rates.

In summary, a tradeoff exists between processor complexity and antenna beam pointing requirements.

Cross-Band and Cross-Polarization Beam Coincidence -- No information is yet available on beam coincidence requirements. Obviously, the closer the coincidence of footprint registration, the more meaningful the cross-band and cross-polarization data becomes. Again, the intended uses of the SMR data must be considered in attempting to qualify beam coincidence requirements. Two major projected uses are: 1) Crop classification, and 2) Soil moisture measurement, with the high frequency being most useful for (1) and the lower frequency most useful for (2). Thus, in specifying the cross-band beam coincidence, one must ask how the data are to be used in an interpretive model. See Section 1.1.

It is conceivable that any static errors could be compensated by increasing the processor complexity. However, dynamic changes in beam coincidence seem to be undetectable and hence uncompensatable.

### 2.3 Conditions Which May Degrade Antenna Performance

The six critical antenna parameters of the last section are directly affected by both random and systematic errors in the electrical excitation and/or mechanical construction and orientation of the SMR array. This section

<sup>†</sup> J.G. Mehlis, Shuttle Synthetic Aperture Radar Implementation Study, Vol. 1, Jet Propulsion Laboratory Document 750-73, Pasadena, California, March 8, 1976, p. 2-42.

discusses the effect of various types of errors upon the performance of the antenna.

### 2.3.1 Electrical Errors

Electrical errors are classified into two types, random and systematic. While both classes will be discussed, it is felt that the effects of systematic errors on SMRA performance will be the more important consideration in the performance of the real antenna since the overall effect of small random errors diminish as the size of the array increases due to an averaging effect. It should be pointed out that random errors do affect the performance of the synthesized antenna pattern (e.g., synthetic beam pointing errors caused by phase decorrelation due to random phase errors), but the extent to which these errors degrade overall system performance cannot be predicted without further study and a knowledge of the image processor operation.

Random electrical errors may be divided into three categories:

1. Amplitude errors.
2. Phase errors.
3. Errors in the individual element patterns.

Random amplitude errors will cause a loss of gain, a broadening of the main beam, a reduction in null depths, and to a lesser extent, a rise in the side lobe level. Random phase errors affect the real beam pointing direction and SLL, as do errors in the characteristics of the individual element patterns. These errors arise from the manufacture-controlled tolerances placed on the antenna which are the result of element-to-element variations in the feed arrangement, power dividers, waveguide slot width and placement, etc. A given random error will affect X-band performance more than L-band performance since phase tolerances are relative to wavelength.

Systematic electrical errors are much more difficult to analyze than random errors since the more important systematic errors depend upon the antenna structure and geometry. Consequently, little a priori information is available for the systematic errors which may influence the performance of the SMRA. The more common (and most easily analyzed) errors are:

1. Linear phase errors.
2. Quadratic phase errors.
3. Cubic phase errors.
4. Periodic errors.

A systematic linear phase shift across the array will change the beam pointing direction, broaden the main beam, decrease the gain, increase the SLL, and degrade polarization purity. Quadratic phase errors tend to raise the null depths between side lobes, widen the main beam, reduce gains, and increase the level of the near-in side lobes. However, the beam direction remains unchanged. Cubic phase errors will shift the beam position, decrease the gain, and produce asymmetrical side lobes. Periodic errors create two additional side lobes and a loss of gain, but no beam pointing errors.

Errors which cannot be classified into one of the above categories must be analyzed separately. The only efficient method of tackling this problem is to simulate the effect of such errors on a digital computer. In this way, information concerning footprint shape, gain, etc., can be obtained for a large class of systematic errors.

The mathematical algorithms for the simulation will be analyzed in Section 3, the organization of the computer program will be discussed in Section 4, and the source listing for the computer program is given in the appendix.

### 2.3.2 Mechanical Errors

Both random and systematic mechanical errors arise from manufacturer-induced tolerances; space environmental effects on the antenna and its supporting structure induce additional systematic departures from flatness. Systematic mechanical errors will likely be more detrimental to SMRA performance than random errors.

The most significant mechanical error problems will be concerned with deviations of the array from nominal flatness. The effect of this error on performance will be the same as a systematic electrical phase error since the change in

element path-length distance introduced by this distortion can be viewed as an electrical phase difference. Hence, a mechanical distortion will exhibit the same effects as the phase distortions of the last section. These include beam defocusing, gain reduction, an SLL increase, and polarization purity degradation.

Another source of mechanical errors is the random and systematic errors in the element locations. Translational errors will change the beam shape, raise the side lobe level, and may introduce other lobes. The effect of rotational errors would be to destroy the polarization purity and to decrease the gain. As with electrical errors, the only efficient method for determining the effect of mechanical errors to devise a suitable computer algorithm. One such algorithm is described in Sections 3 and 4.

### 2.3.3 Other Sources of Errors

One source of error that should not be ignored is the multipath effects caused by reflection from the Shuttle body and/or other experiments. A ray-optics treatment of the problem should be sufficient to identify any possible problems, and to suggest ways of reducing multipath effects to a minimum.

Another consideration to the overall performance of the antenna is the degradation of materials due to the space environment and exhaust envelopes of the Shuttle engines. This could become important for the dielectric used in a microstrip antenna element.

### 2.3.4 Summary of Errors

The effects of various electrical and mechanical errors on SMRA performance is summarized in Table 2.

## 2.4 Summary

The four cause-effect relationships discussed in Section 2 are graphically portrayed in Figure 1. Briefly, manufacturing, environmental, and deployment

TABLE 2. The effects of electrical and mechanical errors on SMRA performance.

Error Type	Gain	Beam Shape	Sidelobe Level	Polarization Purity	Beam Pointing	Beam Coincidence
<u>Electrical</u>						
Random Amplitude	X	X	X			
Random Phase			X		X	X
Linear Phase Shift	X	X	X	X	X	X
Quadratic Phase	X	X	X			
Cubic Phase	X	X			X	X
Periodic Errors	X		X		X	X
<u>Mechanical</u>						
Deviation from Surface Flatness	X	X	X	X	X	X
Translational Errors in Element Location		X	X			
Rotational Errors in Element Location	X			X		

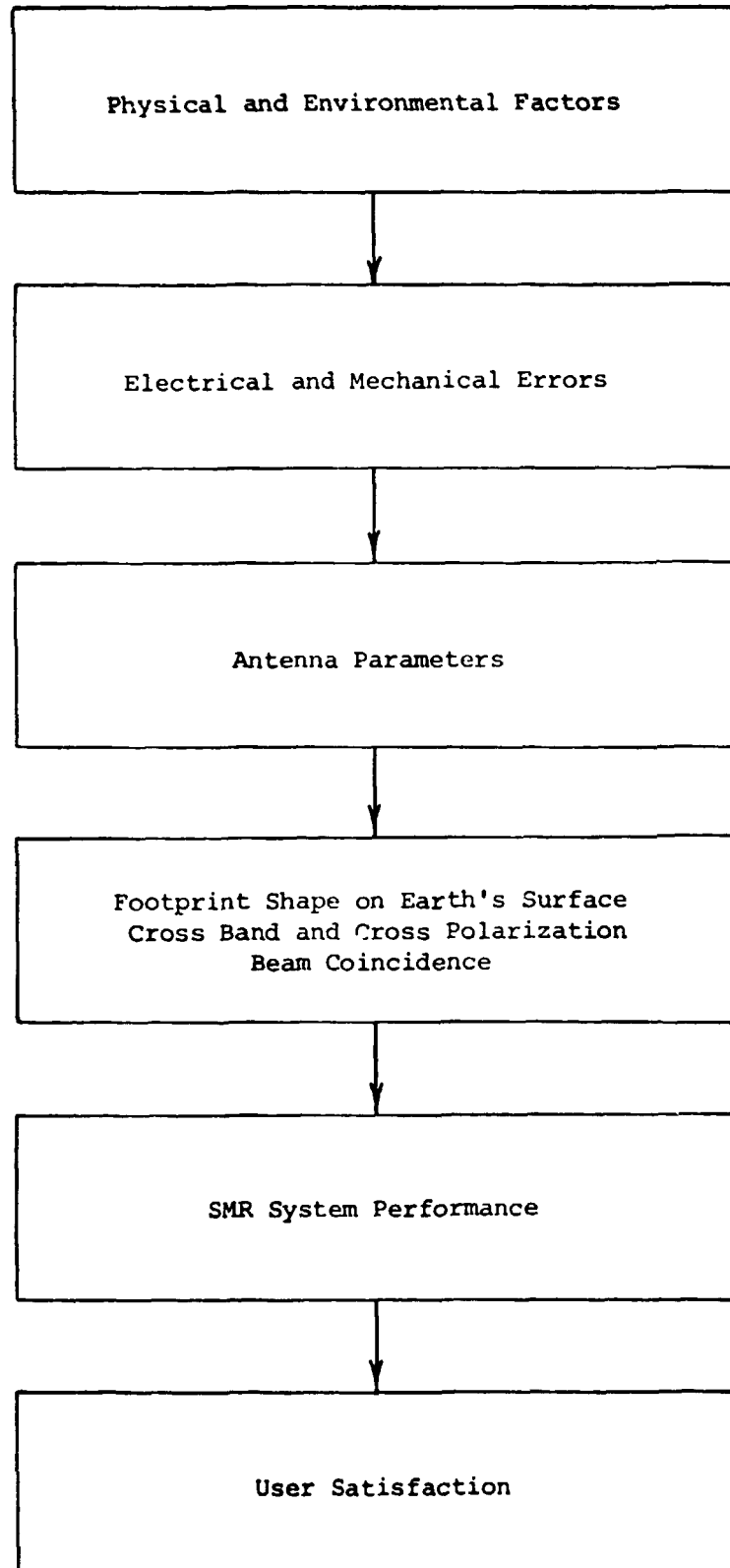


Figure 1. Cause-effect relationships for the SMR Antenna.

factors will create electrical and mechanical errors. These errors in turn affect the performance of the SMR antenna which then changes the characteristics of the ground footprint. Finally, the footprint shape is a major determinant of quality of the overall system.

### 3.0 THE DEVELOPMENT OF A PRELIMINARY SMRA MATHEMATICAL MODEL

To obtain predictions of the behavior of the Shuttle Multispectral Radar Antenna under a wide variety of electrical, mechanical, and environmental conditions, a number of algorithms simulating the SMRA were developed. Emphasis was given to developing mathematical models that would predict the effect of systematic departures from mechanical and electrical flatness on beam pointing accuracy, beam efficiency, etc. Later phases of this work will add to the initial model additional algorithm modules that will take into account the results from thermal, mechanical and electrical tests.

The theory and practical considerations behind the PSL approach to large-scale array modeling was presented in Section 2.2 of the Phase I Interim Report. This method has been expanded upon and implemented in the computer program discussed in Section 4 of this report. The results from a few selected examples are given in Section 5.

#### 3.1 Antenna Electrical and Mechanical Characteristics

A flat uniformly excited rectangular electrically large array of equally-spaced elements lying in the (x, y)-plane will produce a far-field radiation pattern of the form:

$$E = E_0 \frac{\sin M\psi_x/2}{M \sin \psi_x/2} \cdot \frac{\sin N\psi_y/2}{N \sin \psi_y/2} \quad (3-1)$$

where:  $E_0$  = constant

$M$  = number of array elements in azimuth (x)

$N$  = number of array elements in elevation (y)

$\psi_x = \beta d_x u + \alpha_x$

$\psi_y = \beta d_y v + \alpha_y$

$\beta = 2\pi/\lambda$  = wave number

$d_x$  = x-axis interelement spacing

$d_y$  = y-axis interelement spacing

$u$  = cosine of pointing direction with respect to the x-axis

$v$  = cosine of pointing direction with respect to the y-axis



$\alpha_x$  = x-axis interelement phase shift  
 $\alpha_y$  = y-axis interelement phase shift

However, the presence of random and systematic electrical and mechanical errors in the array structure requires a much more general model. Three mechanical and electrical models were considered in the Phase I Interim Report. In essence, each model incorporated an approximation of the electrical and mechanical surfaces. The three varying degrees of approximation are described below.

1. The array factor is the sum of the patterns of each individual element, each of which has its own location, orientation, and excitation. (No approximation)
2. The electrical/mechanical deformation of the array is approximated by a function or sum of functions whose closed-form array factor(s) can be found.
3. The electrical/mechanical deformation of the array is approximated by piecewise bilinear error-minimizing rectangular sections.

The first two of these were either too time-consuming or unable to model adequately severe systematic errors. The third alternative was considered to be the best compromise, especially if the number of subarray sections could be varied to reflect the severity of the distortion.

The geometry of the approximated antenna surface is explained in Figure 2, while Figure 3 shows an individual subarray unit in local coordinates.

### 3.1.1 The Antenna Radiation Pattern Algorithm

Consider an array antenna composed of  $M \times N$  subarray sections. The antenna is nominally located in the  $z=0$  plane and it is centered about the  $z$ -axis as illustrated in Figure 4. The far-field radiation pattern of such an antenna may be written as the weighted sum of the contributions from each subarray.

$$\bar{E}(r, \theta, \phi) = \frac{j 9 \eta}{4 \pi} \frac{e^{-j \beta r}}{r} \sum_{m=1}^M \sum_{n=1}^N a_{mn} e^{j \phi_{mn}} \bar{g}_{mn}(u, v) \exp [j \psi]$$

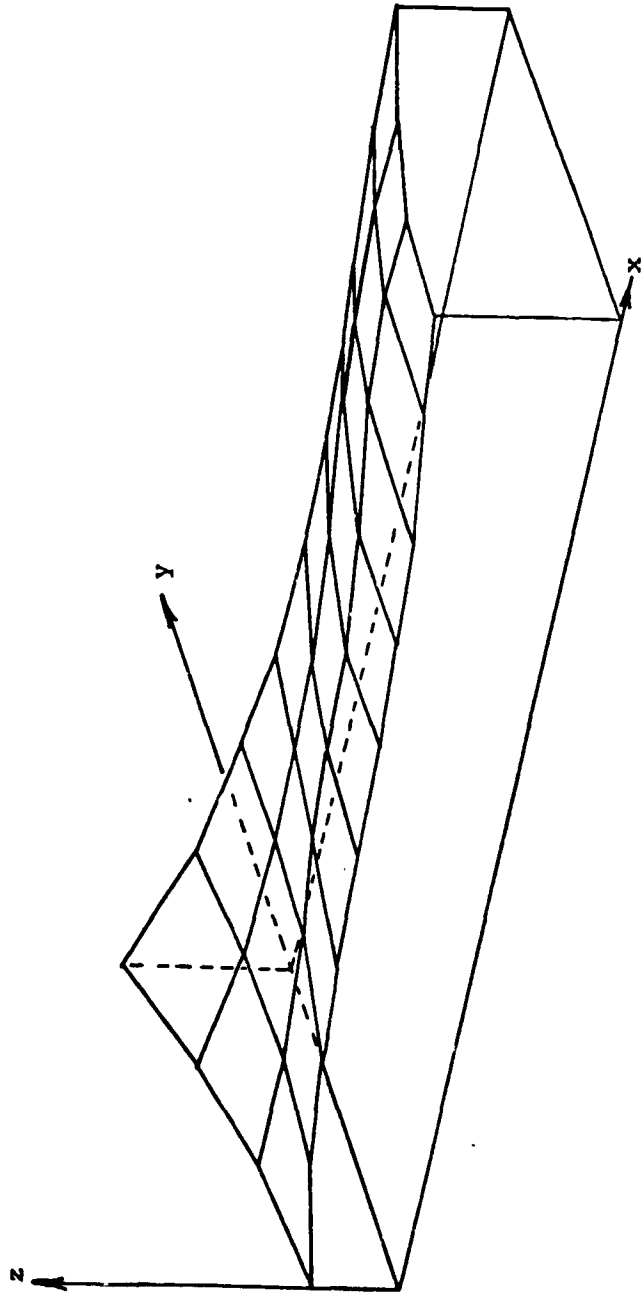


Figure 2. Geometry of the approximated antenna surface.

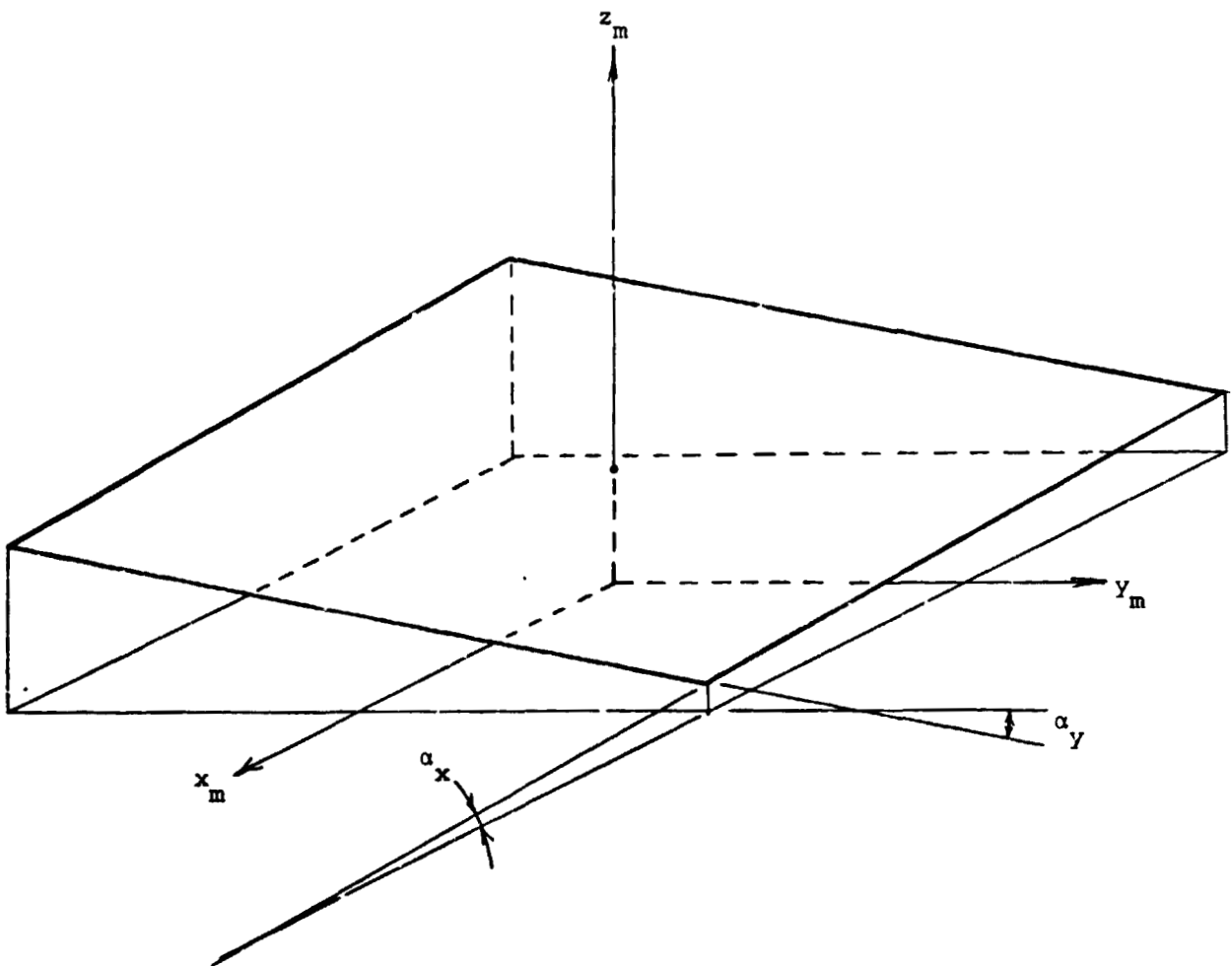


Figure 3. Single subarray section in local coordinates.

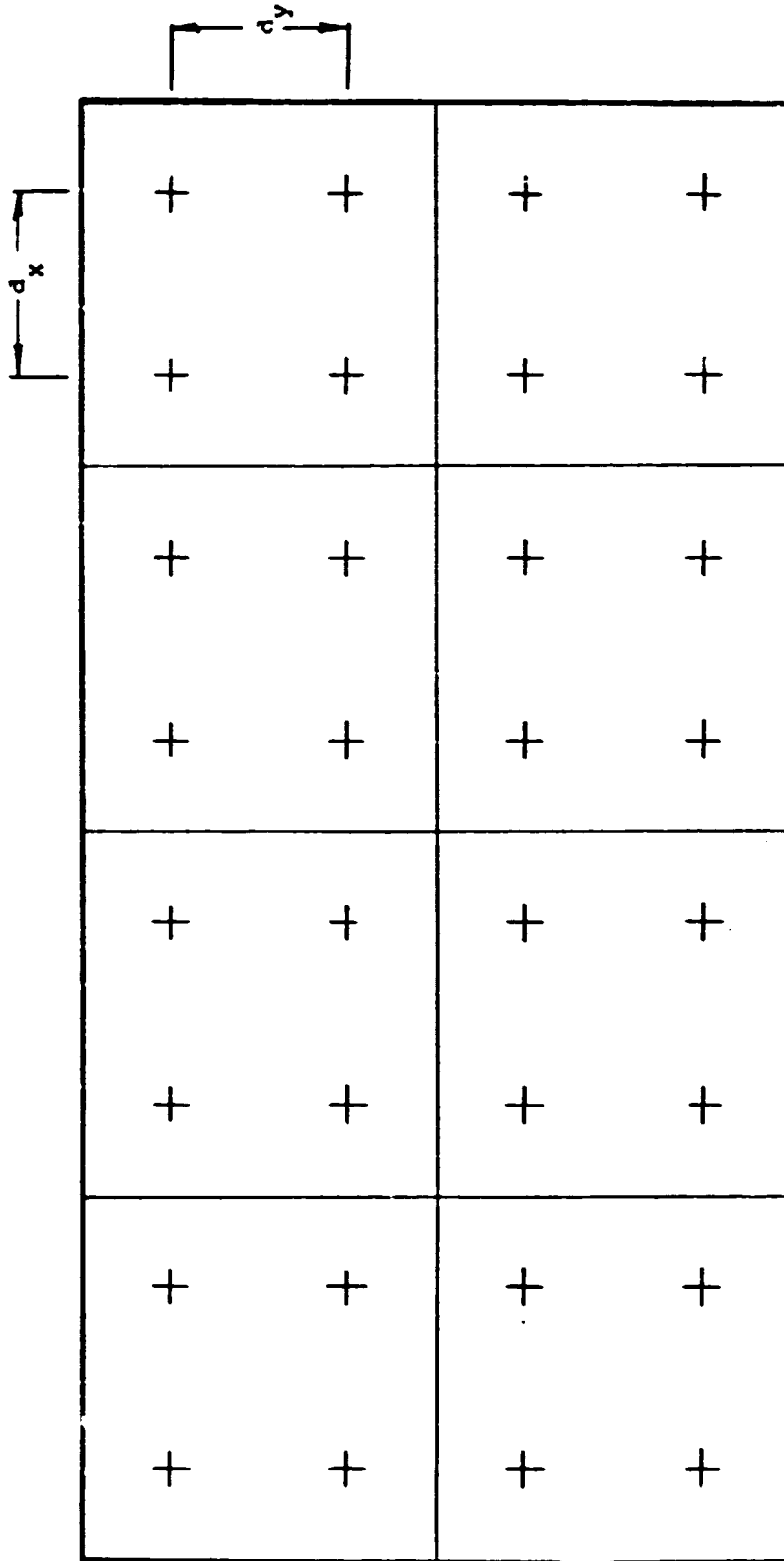


Figure 4. An array antenna of 8 x 4 elements divided into 4 x 2 subarrays. The element spacings  $d_x$  and  $d_y$  are indicated.

where:  $a_{mn} e^{j\phi_{mn}}$  = the complex excitation of the  $mn^{\text{th}}$  subarray.  
 $(u, v, w)$  =  $(\sin\theta\cos\phi, \sin\theta\sin\phi, \cos\theta)$ , the cosines of the beam pointing direction with respect to the x, y, and z axes.  
 $\beta = 2\pi/\lambda$  = wave number of the source.  
 $\eta = \sqrt{\frac{\mu}{\epsilon}}$  = intrinsic impedance of the medium.  
 $\bar{g}_{mn}(u, v)$  = vector array factor of the  $mn^{\text{th}}$  subarray.  
 $\psi = (x_{mn}u + y_{mn}v + z_{mn}w)$   
 $(x_{mn}, y_{mn}, z_{mn})$  = location of the center of the  $mn^{\text{th}}$  subarray.

For rectangular sections with dipole elements, the vector "subarray factor" has the form:

$$\bar{g}_{mn}(u, v) = \frac{\sin N_x \psi_x / 2}{N_x \sin \psi_x / 2} \frac{\sin N_y \psi_y / 2}{N_y \sin \psi_y / 2} \begin{cases} \frac{\cos \frac{\pi}{2} u}{\sqrt{1-u^2}} & \hat{x} \text{ Horizontal Polarization} \\ \frac{\cos \frac{\pi}{2} v}{\sqrt{1-v^2}} & \hat{y} \text{ Vertical Polarization} \end{cases} \quad (3-3)$$

where:  $N_x, N_y$  = number of elements in the subarray.  
 $\psi_x = \beta d_x \cos(\cos^{-1}u - \alpha_x) + \phi_x$   
 $\psi_y = \beta d_y \cos(\cos^{-1}v - \alpha_y) + \phi_y$   
 $d_x, d_y$  = interelement spacing.  
 $\phi_x, \phi_y$  = interelement phase shift.  
 $\alpha_x, \alpha_y$  = the x-axis and y-axis angular tilt of the section.

The (m,n) subscripts have been dropped for clarity.

Notice that for a given linear polarization,  $|\bar{g}_{mn}|$  may be written as:

$$|\bar{g}_{mn}(u, v)| = f_1(u) f_2(v)$$

This separability allows one to precalculate  $f_1(u)$  and  $f_2(v)$  so that  $\bar{g}_{mn}$  may be calculated using a table look-up algorithm, greatly reducing the numbers of operations and function calculations. Since all subarray

sections are identical except for orientation, the same  $f_1$  and  $f_2$  tables may be used for each. The total far-field radiation pattern is then obtained by vectorally summing the contributions from each subarray.

### 3.2 Antenna Position and Orientation

The spacecraft position and orientation are specified by its altitude and three angles. These angles (YAW, TILT, and TWIST) are defined by Figure 5. While it is true that only two angles are needed to specify any possible (static) orientation of the antenna, three angles are used to enable the user to picture clearly the antenna pointing direction. The antenna subsatellite point is assumed here to be  $0^\circ$  Longitude,  $0^\circ$  Latitude as shown in Figure 6.

The matrices for the three rotations are:

$$\text{YAW: } \begin{bmatrix} x' \\ y' \\ z' \end{bmatrix} = \begin{bmatrix} \cos(\text{YAW}) & \sin(\text{YAW}) & 0 \\ -\sin(\text{YAW}) & \cos(\text{YAW}) & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} x \\ y \\ z \end{bmatrix} \quad (3-4)$$

$$\text{TILT: } \begin{bmatrix} x'' \\ y'' \\ z'' \end{bmatrix} = \begin{bmatrix} 1 & 0 & 0 \\ 0 & \cos(\text{TILT}) & -\sin(\text{TILT}) \\ 0 & \sin(\text{TILT}) & \cos(\text{TILT}) \end{bmatrix} \begin{bmatrix} x' \\ y' \\ z' \end{bmatrix} \quad (3-5)$$

$$\text{TWIST: } \begin{bmatrix} x''' \\ y''' \\ z''' \end{bmatrix} = \begin{bmatrix} \cos(\text{TWIST}) & \sin(\text{TWIST}) & 0 \\ -\sin(\text{TWIST}) & \cos(\text{TWIST}) & 0 \\ 0 & 0 & 1 \end{bmatrix} \begin{bmatrix} x'' \\ y'' \\ z'' \end{bmatrix} \quad (3-6)$$

In practice, the YAW, TILT, and TWIST matrices would be multiplied together and stored as a single 3x3 matrix.

### 3.3 Coordinate Transformation To Produce The "Footprint" On The Earth's Surface

There are two transformations involved:

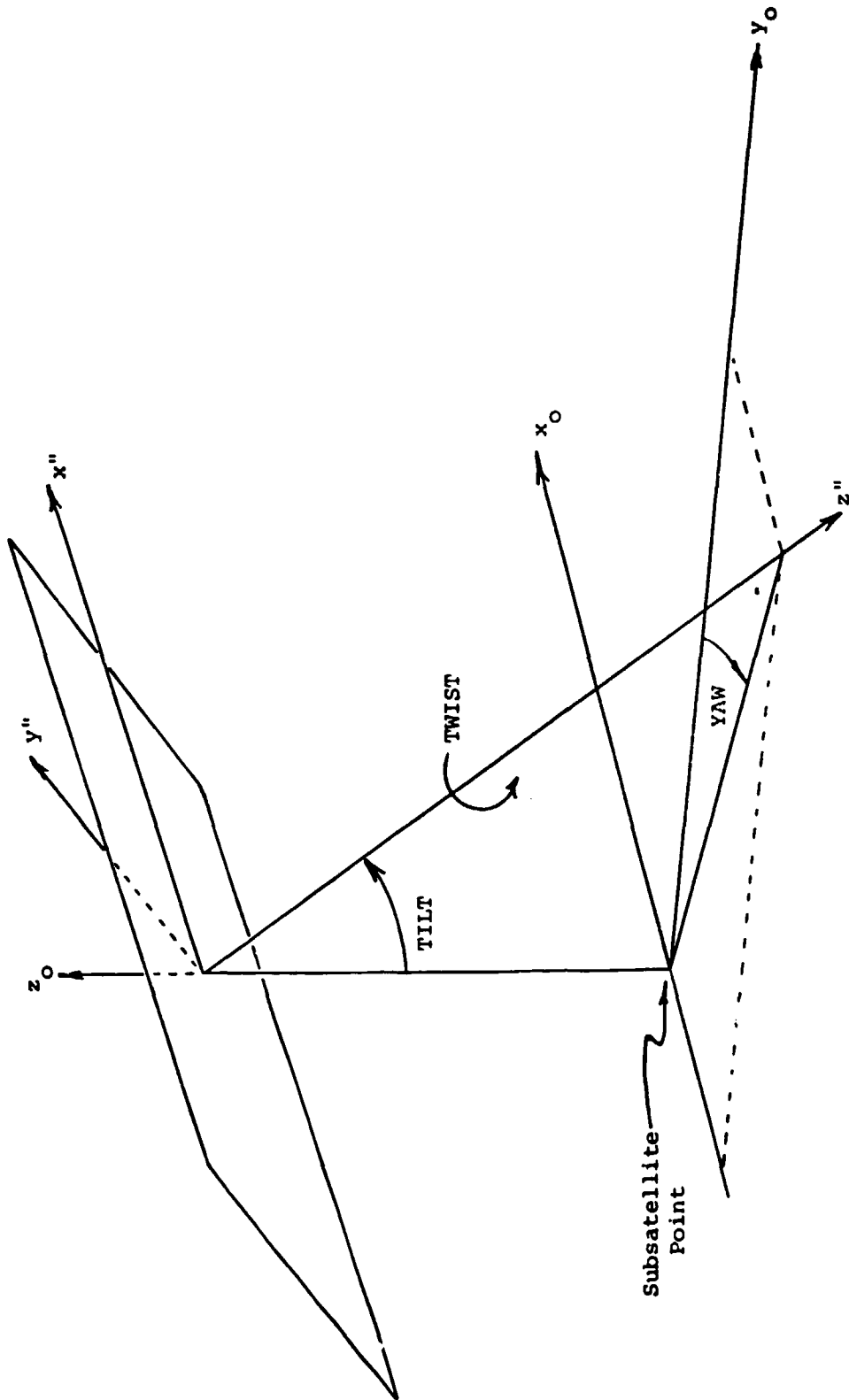


Figure 5. Orientation angles and geometry for the SMR antenna.

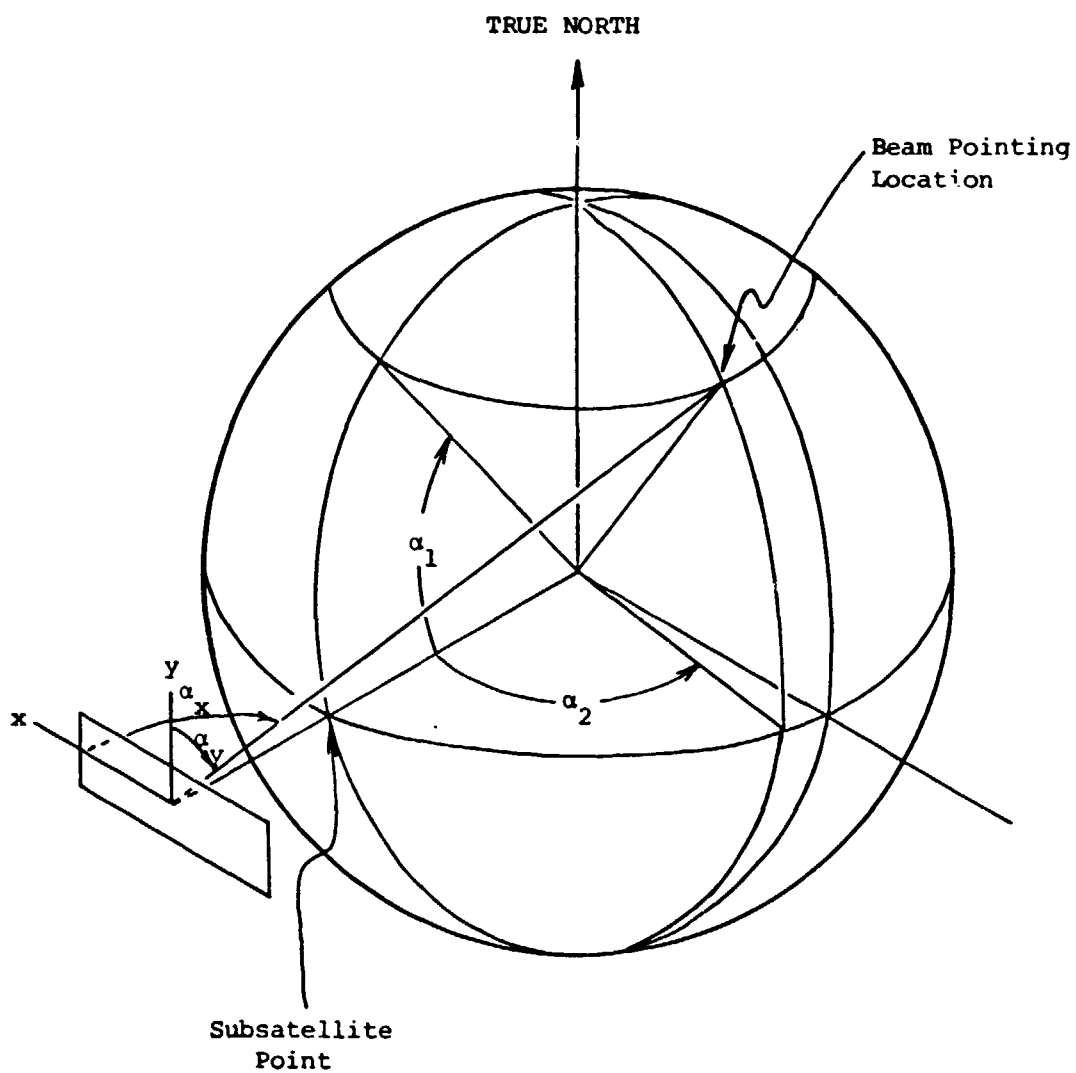


Figure 6. SMRA antenna above a spherical earth. The subsatellite point is at  $0^\circ$  latitude,  $0^\circ$  longitude. The beam is pointed at  $[\alpha_1(\text{latitude}), \alpha_2(\text{longitude})]$ .



- (1) What point, i.e., with latitude ( $\alpha_1$ ) and longitude ( $\alpha_2$ ), on the earth's surface corresponds to a given beam direction ( $u, v$ )?
- (2) What antenna beam direction ( $u, v$ ) corresponds to a specified latitude and longitude location?

The first transformation is used to predict the location of the beam center. The second is necessary to determine the "footprint" contours of the antenna.

### 3.3.1 Latitude and Longitude From Beam Direction

For an unrotated antenna pointing to a nadir of ( $0^\circ, 0^\circ$ ), the following formulas may be derived:

$$\alpha_r = \sin^{-1} \left[ \frac{R_e + \text{Alt}}{R_e} \right] \sin \theta - \theta \quad (\alpha_r > 0)$$

$$\phi_r = \phi - \frac{\pi}{2} \quad (3-7)$$

$$\text{Range} = R_e \cdot \alpha_r$$

where:  $R_e$  = radius of the (spherical) earth.  
 $\text{Alt}$  = antenna altitude in same units as  $R_e$ .  
 $(\theta, \phi)$  = pointing direction of unrotated antenna.  
 $\alpha_r$  = range angle.  
 $\phi_r$  = heading angle.

$$\text{Then: } \alpha_2 = \tan^{-1} (\sin \phi_r / \tan \alpha_r)$$

$$\alpha_1 = \cos^{-1} (\cos \alpha_r / \cos \alpha_2) \quad (3-7a)$$

If  $\alpha_1 > 90^\circ$ ,  $\alpha_1 = \alpha_1 - 180^\circ$ .

Example, let  $R_e = 6400$  km,  $\text{Alt} = 200$  km, and  $(\theta, \phi) = (60^\circ, 120^\circ)$ . Then:

$$\alpha_r = 3.263975^\circ \quad \phi_r = 30^\circ$$

$$\alpha_2 = 1.6333^\circ \quad \alpha_1 = 2.8263^\circ$$

$$\text{Range} = 364.56 \text{ km.}$$

### 3.3.2 Beam Direction From Latitude and Longitude

Given  $\alpha_1$  and  $\alpha_2$  we may invert eqns. (3-7a) to find:

$$\begin{aligned}\alpha_r &= \cos^{-1}(\cos\alpha_1 \cdot \cos\alpha_2) \\ \phi_r &= \sin^{-1}(\cos\alpha_1 \cdot \cos\alpha_r \cdot \sin\alpha_2)\end{aligned}\quad (3-8)$$

Then:

$$\begin{aligned}\phi &= \phi_r + 90^\circ \\ z &= (R_e + \text{Alt}) - R_e \cos\alpha_1 \cos\alpha_2 \\ x &= -R_e \cos\alpha_1 \sin\alpha_2 \\ y &= R_e \sin\alpha_1 \\ r &= \sqrt{x^2 + y^2 + z^2} \\ \theta &= \cos^{-1}(z/r)\end{aligned}\quad (3-9)$$

Example, let  $R_e = 6400$ ,  $\text{Alt} = 200$  km,  $\alpha_1 = 2.82631^\circ$ , and  $\alpha_2 = 1.6333^\circ$ . Then:

$$\begin{aligned}\alpha_r &= 3.263975^\circ & x' &= -182.19476 \text{ km.} \\ \phi_r &= 29.9997 \approx 30^\circ & y' &= 315.5738 \text{ km.} \\ \phi &= 120^\circ & r' &= 420.764 \text{ km.} \\ z' &= 210.382 \text{ km.} & \theta &= 60^\circ\end{aligned}$$

### 3.4 Data Handling Techniques and Data Reduction

With the vast amounts of data generated by this algorithm, it was clear that in order to obtain usable results, the data must be reduced to pictorial form. Five modes of presentation were developed. These are:

1. Printer profile plots of "principal planes."
2. Printer contour plot of the entire region of interest.
3. Plotter profile plots of "principle planes."
4. Plotter contour plots of the entire region.
5. Plotter three-dimensional plots of the entire region.

The printer plots provide a quick and inexpensive preview of the results of the simulation. If satisfactory results are obtained, then the Calcomp plotter is used to draw more detailed plots.

When the Tektronix 4051 interactive graphics terminal is made operational, many of the data presentation problems will be greatly simplified. It will be then possible to quickly draw a complete contour "footprint" from any data set. If a permanent copy is desired, a hard copy or photograph may be taken.

### 3.5 Areas For Further Development

The mathematical model discussed here is preliminary in nature and its algorithms will be refined and expanded during Phase II.

The most significant addition to the model will be the incorporation of data gathered from near-field and far-field electrical measurements, thermal measurements, and mechanical tests of the subarray panels. These data, to be taken during Phase IV, will be used by the model to predict the performance of the full-size SMR antenna. It is planned to add these modules to the algorithm during Phase III.

The other important addition to the model will be the inclusion of a module to accept arbitrary array elements and/or total antenna type. With this change, other antenna types, such as reflector antennas, other aperture antennas, and traveling wave antennas, can be simulated as well as arrays of antennas with arbitrary elements.

### 3.6 Summary and Block Diagram Of The Simulation Algorithm

The operation of the SMRA mathematical model can be summarized by the block diagram shown in Figure 7. Only those modules presently implemented are shown.

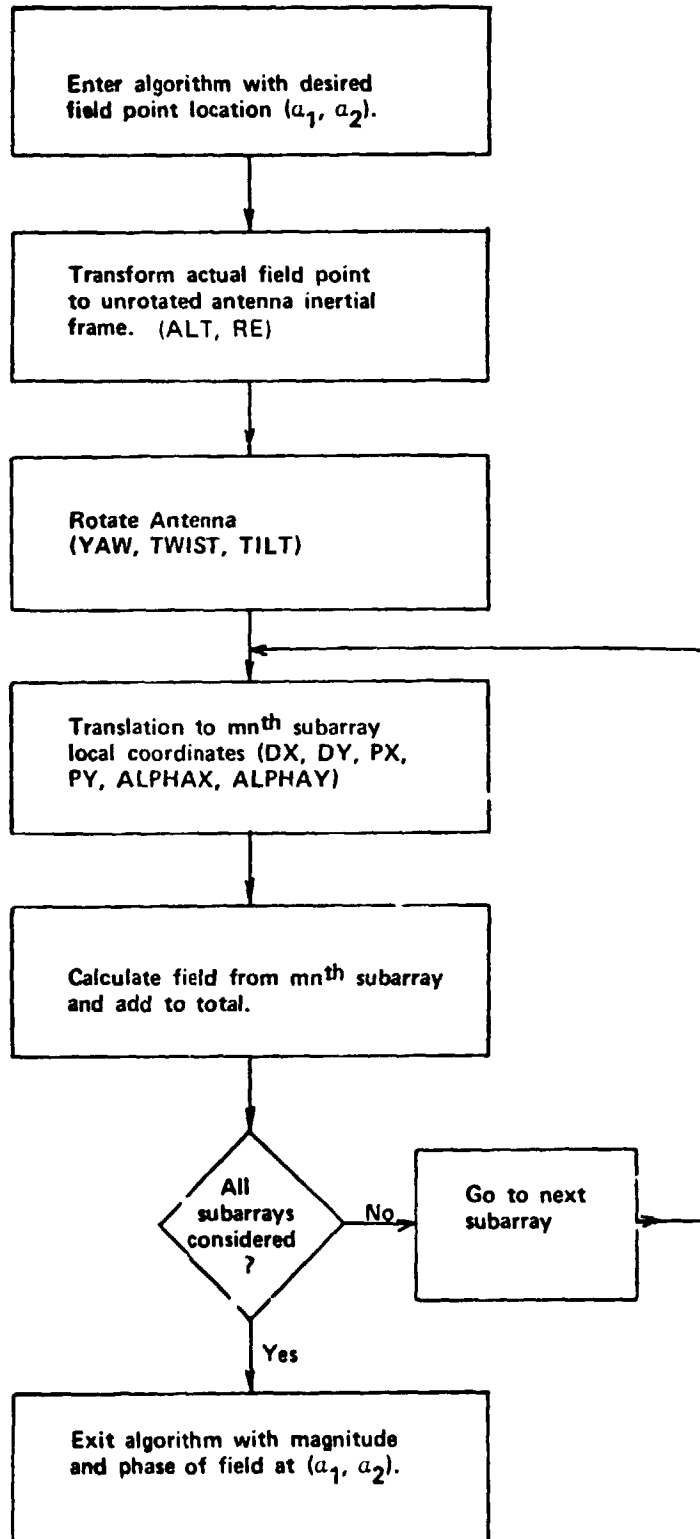


Figure 7. Operation of the SMRA mathematical model.

#### 4.0 THE SMRA SIMULATION PROGRAM

A computer program was written to implement the simulation algorithms of Section 3. It produces line printer plots and Calcomp plotter plots of the antenna footprint on the earth's surface. The organization of the program is shown in Figure 8, and a description of each program segment is given in Section 4.1.

##### 4.1 Program Description

- MAINPGM - Performs mostly housekeeping chores and calls other routines according to user input commands. It calculates the "YAW-TILT-TWIST" array, predicted beam center location, plot normalization factor (pattern value at predicted beam center location), increments beam pointing angle to cover entire "footprint" region and initiates profiles along lines of constant latitude and longitude intersecting the predicted beam center.
- ANTENA - Inputs and calculates appropriate antenna parameters: number of elements, spacing, phase shift, number of subarrays, polarization, and element type. After calculations have been performed, a summary is printed.
- MECH - Calculates mechanical deformation data based on the inputs from STRESS, THERML, and MISC. For each subarray, MECH calculates the average displacement ZAVG, the tilt angles, ALPHAX and ALPHAY, and the error coefficient of the bilinear approximations. A summary is printed after execution.
- STRESS - Determines the warpage of the array that is due to mechanical considerations.
- THERML - Will determine the warpage of the array due to thermal gradients across and through the array structure. (Not Yet Implemented)

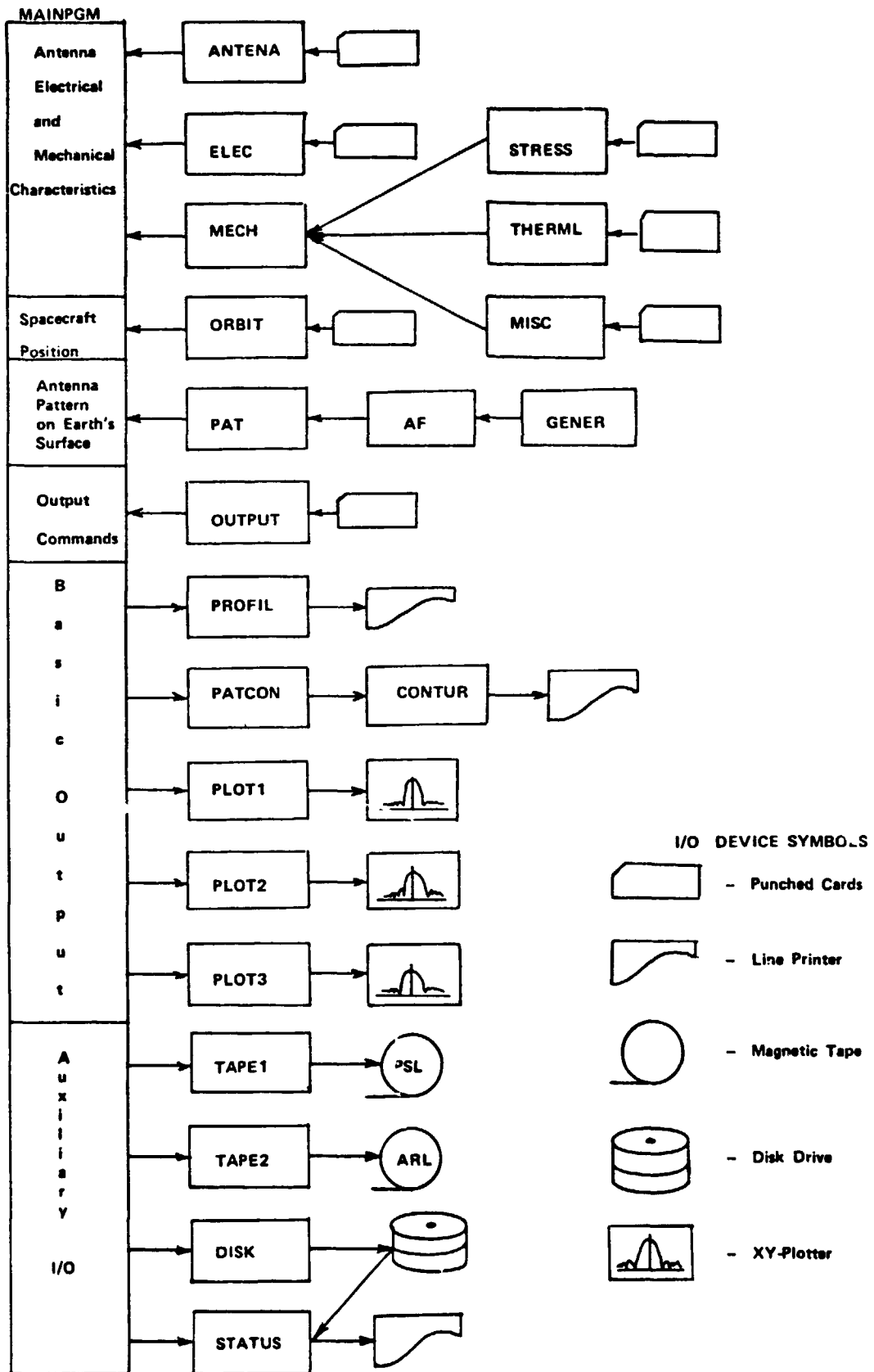


Figure 8. Organization of the SMRA Simulation Program.

- MISC - Inputs any other deformation data not covered by STRESS and THERML.
- ELEC - Inputs the amplitude, phase, and linear phase gradients of each subarray. This information is passed to PAT via common block ELCTRC, and a summary is printed.
- ORBIT - Inputs orbital parameters and antenna YAW, TILT, and TWIST, then prints a summary.
- PAT - Is the basic pattern subprogram. Given latitude-longitude coordinates on the earth's surface, it performs all translations and rotations to determine the beam pointing angle (u,v). It then calls AF to find the subarray factor and finally computes the sum of all subarrays.
- AF - This function calculates the array factors of a subpanel by "table look-up" with linear interpolation if  $|u| < UMAX$  and  $|v| < VMAX$ . Otherwise, the AF is computed in the usual way.
- GENER - This routine generates the table used by AF. Table length is 1001 words: UMAX and VMAX are chosen to include the main beam plus the first three side lobes.
- OUTPUT - Translates user commands for printer and pattern output into logical switches for the program.
- PROFIL - Is the printer profile plot routine. It will print a one-dimensional plot down the page for up to 501 data points. The exact values of both the abscissa and ordinate are printed with each data point, the ordinate also being printed in dB.
- PATCON - Is chiefly a "bookkeeping" subroutine for CONTUR. It calls subroutine CONTUR three times to generate a complete 151 x 151 two-dimensional contour map.

- CONTUR - Prints a 51 x 151 contour plot of the footprint. To obtain a complete 151 x 151 plot, this routine is called three times by subroutine PATCON. Then the three separate plots are pasted together to make the composite.
- PLOT1 - Is the Calcomp one-dimensional plot routine. It is called twice by the main program, and it is used to generate the profile plots of the principal planes through the main beam.
- PLOT2 - Is the Calcomp contour map routine. From the data used to generate the printer contour map, it draws a continuous contour plot of the antenna ground footprint over the rectangular region specified by the user.
- PLOT3 - Is the Calcomp three-dimensional plot routine. It plots the magnitude (in dB) of the footprint pattern over rectangular region specified by the user.
- TAPE1 - Will be used to store the pattern matrices and other important variables so that additional plots may be generated by other programs without recomputing the data. (Not Yet Implemented)
- TAPE2 - Is the ARL data tape routine. It stores the antenna pattern as a function of u and v and is used whenever ARL requires a new antenna pattern for their processor modeling program. (At present, TAPE2 is a separate computer program.)
- DISK - Will be used to store pattern data on a disk drive at PSL. (Not Yet Implemented)
- STATUS - Will be used to determine what data has been stored by TAPE1 and DISK, to selectively erase any simulation run, and to reinitialize the entire storage area. (Not Yet Implemented)



## 4.2 Input/Output and Flexibility To The User

The program input data has been structured so that user knowledge of computer fundamentals is not required. Consequently, the program can be used by anyone desiring information concerning the footprint of a particular antenna configuration.

The input has been divided into six modules. These are:

1. Simulation Information
2. Antenna Position and Orientation
3. Output Commands and Parameters
4. Antenna Configuration
5. Antenna Mechanical Parameters
6. Antenna Electrical Parameters

By including input parameters in these six categories, it is possible for the user to simulate a wide range of electrical, mechanical, and physical scenarios. Furthermore, a change in the simulation does not require that the complete input deck be repunched. When related scenarios are being studied, only one of the six modules need be changed.

Examples of different simulations are deferred to Section 5.

### 4.2.1 Simulation Information

The simulation information consists of three entries: 1) the simulation number, 2) the date of the simulation, and 3) a short narrative describing the simulation. The simulation number and date provide bookkeeping reference information. The narrative is printed at the beginning of the computer output to allow quick identification of the program output and future dates.

### 4.2.2 Antenna Position and Orientation

The parameters used as input for this category are antenna altitude, yaw, tilt, twist, and frequency. Altitude is expressed in kilometers, the angles

are expressed in degrees, and frequency is expressed in GHz. These parameters are more fully described in Section 3.2.

#### 4.2.3 Output Commands and Parameters

Five logical switches are used to control the five output categories of Section 3.4. If a particular output is desired, the associated command switch is set to logical "1". In addition to the output commands, it is necessary to specify parameters for those output devices which have been turned on by the output commands.

The parameters needed for a particular simulation are governed by which logical switches have been previously engaged. If any output command is entered, then the latitude and longitude limits of the footprint must be entered. If a plotter contour map is desired the contour level parameters are required. For a printer contour, the highest contour, lowest contour, and contour level interval must be entered. For any two- or three-dimensional plot, the plot resolution is necessary.

Some outputs do not require a command. The narrative described above and the summaries described in Section 3 have no command word; this information is always printed.

#### 4.2.4 Antenna Configuration

This category contains all the parameters needed by subroutine ANTENA. These include the number of subarray sections, the number of elements, the interelement spacings in centimeters, the interelement phase shifts in degrees, the element type, and the element polarization.

#### 4.2.5 Antenna Mechanical Information

Antenna mechanical information includes all the parameters needed by subroutines STRESS, THERML, and MISC. In this preliminary simulation model, the displacement of the antenna from nominal flatness (expressed in centimeters) is entered through subroutine MISC.

#### 4.2.6 Antenna Electrical Parameters

All excitation information is entered through subroutine ELEC. The necessary parameters provide information concerning the excitation magnitude, phase, and any linear phase taper for each subarray section.

## 5.0 EXAMPLES OF COMPUTER SIMULATION RUNS

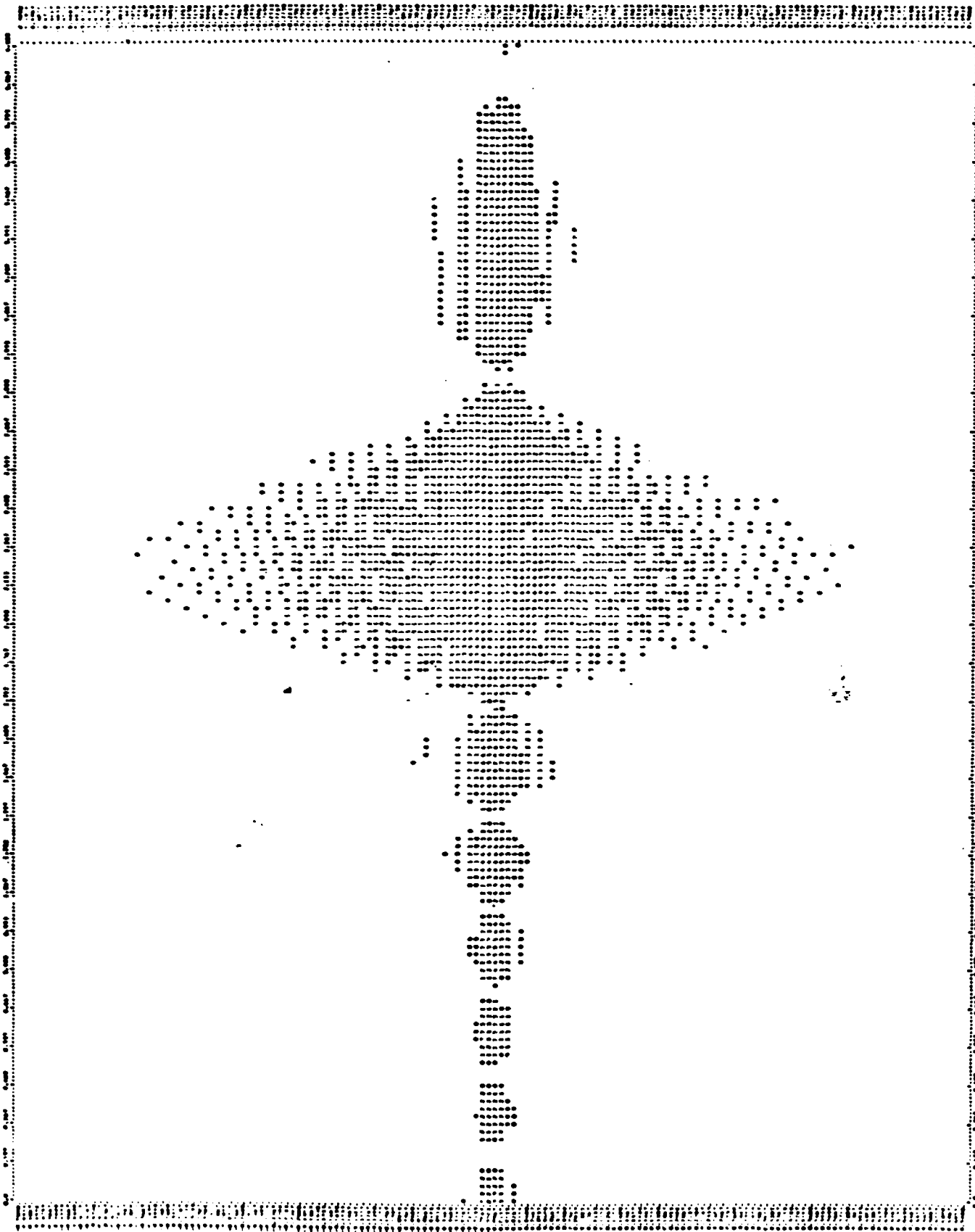
The following pages illustrate the actual line printer output from the Shuttle Multispectral Radar Antenna Simulation Program. The pages run in sequence; no attempt has been made to join the profile plot or contour map pages together. However, a photo-reduction of a typical composite contour map is shown in Figure 9.

The comments printed with the pattern output provide all the necessary information to visualize each scenario being simulated. All antennas have the following characteristics in common:

1. Frequency = 9.0 GHz
2. Tilt Angle =  $50^{\circ}$
3. Altitude = 200 km
4. Polarization: Horizontal
5. Configuration: Both modules excited for minimum beam width
6. Number of elements: 504 x 12
7. Element spacing: 2.2966 x 2.3550 cm
8. Phase Shift:  $0^{\circ}$

Pattern 100 is the simulation of the baseline design, with no electrical or mechanical errors. Pattern 101 illustrates pattern errors due to misalignment of the three panels. Pattern 102 shows the effect of a spherical bow in the antenna which might be induced by thermal gradients through the antenna surface. A systematic phase error is simulated by Pattern 103, and Pattern 104 provides information on errors caused by Shuttle attitude errors.

REPRODUCIBILITY OF THE  
ORIGINAL PAGE IS POOR



PATTERN 100

SPACE SHUTTLE IMAGING RADAR ANTENNA SIMULATION PROGRAM

25 OCTOBER 1976

EXAMPLE 1:  
 BASELINE DESIGN FOR X-RANG SPACE SHUTTLE IMAGING RADAR (90TH PANELS).

SYSTEM INFORMATION:  
 FREQUENCY = 9.000 GHz.  
 YAW = 0.0 DEGREES  
 TILT = 50.000 DEGREES  
 TWIST = 0.0 DEGREES  
 ALTITUDE = 200.000 KM.

ANTENNA PARAMETERS FOR SIMULATION NUMBER 100  
 ELEMENT TYPE: HORIZONTAL CIRCLE  
 NUMBER OF ELEMENTS (X,Y) = (504 , 12)  
 INTERELEMENT SPACING (CM.): 2.2966 , 2.3550  
 INTERELEMENT PHASE SHIFT (DEG.): 0.0 , 0.0

DEFORMATION DATA FOR SIMULATION 100:

0.0 0.0

0.0 0.0

ELECTRICAL DATA FOR SIMULATION 100:

( 1, 1 ) ( 1  
 PHSX: 0.0  
 PHSY: 0.0  
 AMAG: 1.00  
 APHS: 0.0

PRINT/PLOT INFORMATION:  
 REQUESTED OUTPUT:  
 PRINTER PROFILE  
 PRINTER CONTROL  
 PLOT RESOLUTION: 151 X 151 POINTS

STARTX = -0.500  
 STOPX = 0.500  
 DELTAX = 0.007

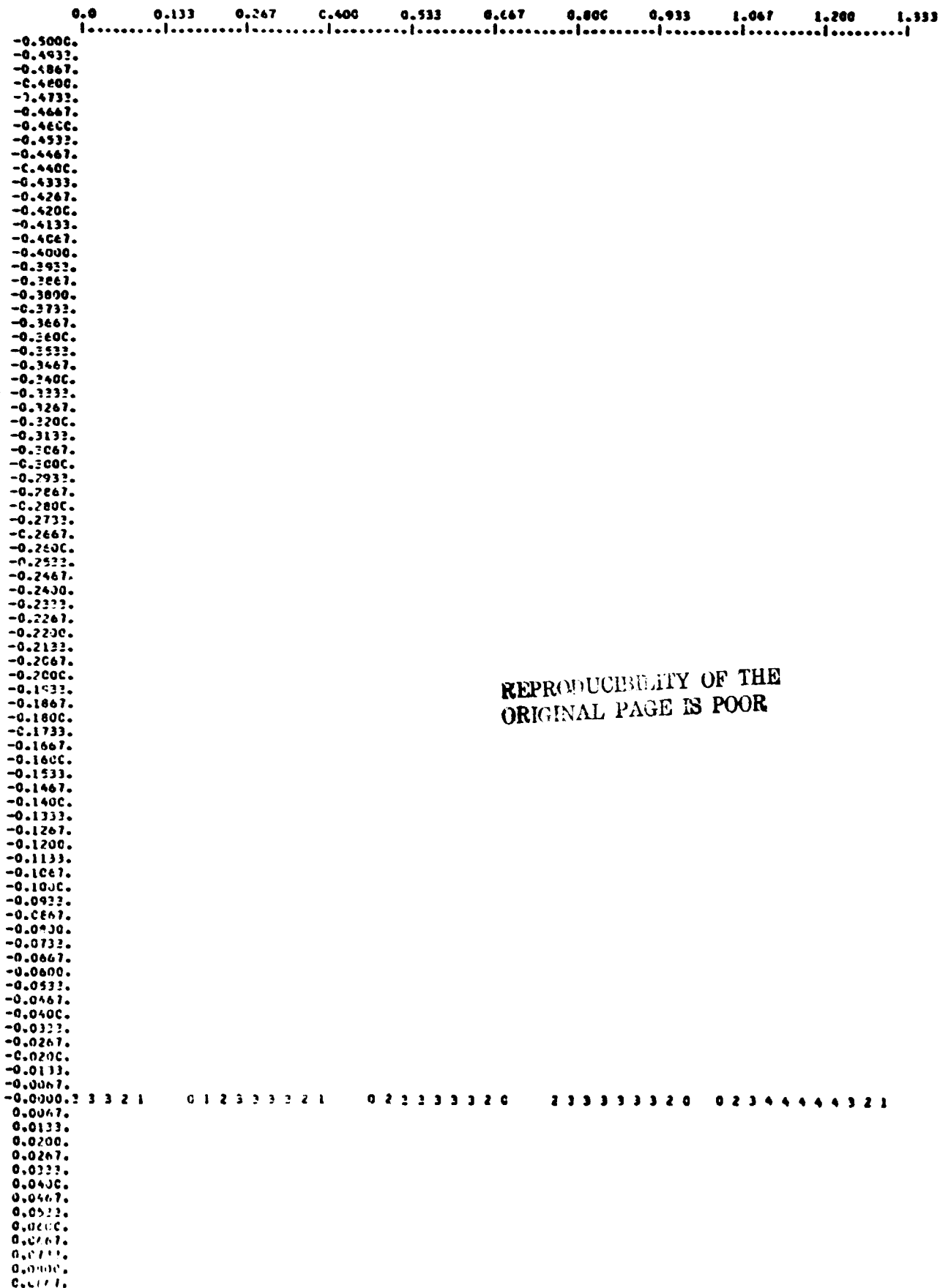
STARTY = 0.0  
 STOPY = 4.000  
 DELTAY = 0.027

AREA	XCENT	YCFNT	ZAVG	ALPMAX	ALPMAY	AMAG	APMS	PHSX	PMSY
1	0.0	0.0	0.0	0.0	0.0	1.0000	0.0	0.0	0.0

PREDICTED BEAM CENTER:  
 LATITUDE = 2.1645  
 LONGITUDE = 0.0  
 RANGE = 200.000  
 BEARING = 0.0

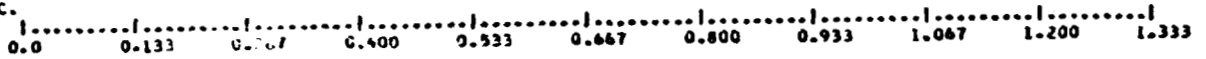
PLCT NORMALIZATION FACTOR = -35.611 C9.

PRINTER CONTROL PLOT FOR SIMULATION NUMBER 100



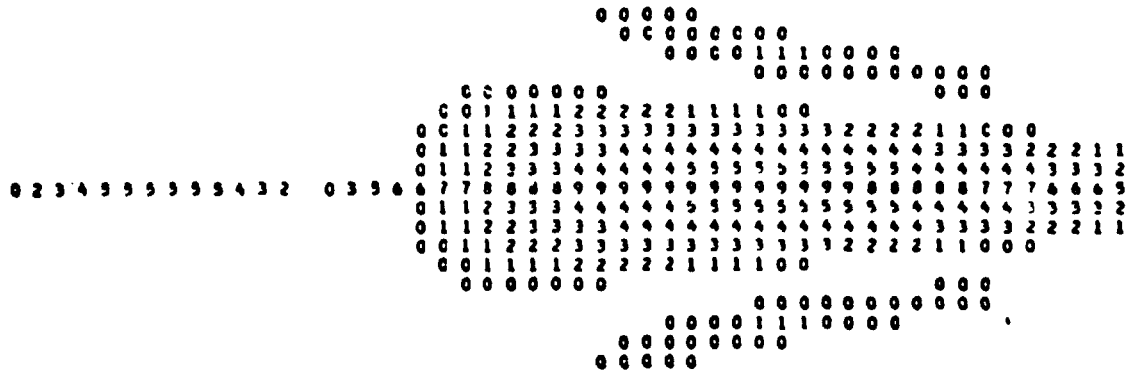


0.0933.  
0.1000.  
0.1067.  
0.1133.  
0.1200.  
0.1267.  
0.1333.  
0.1400.  
0.1467.  
0.1533.  
0.1600.  
0.1667.  
0.1733.  
0.1800.  
0.1867.  
0.1933.  
0.2000.  
0.2067.  
0.2133.  
0.2200.  
0.2267.  
0.2333.  
0.2400.  
0.2467.  
0.2533.  
0.2600.  
0.2667.  
0.2733.  
0.2800.  
0.2867.  
0.2933.  
0.3000.  
0.3067.  
0.3133.  
0.3200.  
0.3267.  
0.3333.  
0.3400.  
0.3467.  
0.3533.  
0.3600.  
0.3667.  
0.3733.  
0.3800.  
0.3867.  
0.3933.  
0.4000.  
0.4067.  
0.4133.  
0.4200.  
0.4267.  
0.4333.  
0.4400.  
0.4467.  
0.4533.  
0.4600.  
0.4667.  
0.4733.  
0.4800.  
0.4867.  
0.4933.  
0.5000.



PRINTER CONTCUP PLOT FOR SIMULATION NUMBER 100

1.333 1.467 1.600 1.733 1.867 2.000 2.133 2.267 2.400 2.533 2.667  
|.....|.....|.....|.....|.....|.....|.....|.....|.....|.....|









Z-AXIS PROFILE FLCT ALONG 2.194 DEGREES LATITUDE.  
PATTERN NUMBER 100

SCALE FACTOR IS 10<sup>-2</sup>

DB.	REAL	99.9983	79.9988	59.9993	39.9998	20.0003	0.0008
-46.59	C.468						0.5000
-45.83	0.511						0.4980
-50.60	0.295						0.4960
-62.74	0.073						0.4940
-47.81	0.407	*	*	*	*	*	0.4920
-45.41	0.537						0.4900
-48.08	0.294						0.4880
-65.99	0.050						0.4860
-49.82	0.323						0.4840
-45.45	0.534						0.4820
-46.47	C.475						0.4800
-55.30	0.172						0.4780
-53.10	0.221	*	*	*	*	*	0.4760
-45.94	C.505						0.4740
-45.48	0.522						0.4720
-50.84	C.287						0.4700
-59.39	0.107						0.4680
-46.94	C.450						0.4660
-44.96	0.565						0.4640
-48.17	0.390						0.4620
-77.03	C.014	*	*	*	*	*	0.4600
-48.56	C.373						0.4580
-44.86	0.572						0.4560
-46.42	C.478						0.4540
-57.29	0.137						0.4520
-51.12	C.278						0.4500
-45.15	0.553						0.4480
-45.27	0.545						0.4460
-51.87	0.255	*	*	*	*	*	0.4440
-55.44	0.169						0.4420
-45.25	C.510						0.4400
-44.52	C.590						0.4380
-48.76	0.365						C.4360
-65.78	0.051						0.4340
-47.03	0.444						0.4320
-44.27	0.611						0.4300
-46.72	0.462	*	*	*	*	*	0.4280
-62.98	C.071						0.4260
-48.85	0.361						0.4240
-44.31	C.609						0.4220
-45.31	0.542						0.4200
-54.21	0.193						0.4180
-51.66	C.261						0.4160
-44.68	0.583						0.4140
-44.38	C.604	*	*	*	*	*	0.4120
-50.19	C.310						0.4100
-56.48	C.150						0.4080
-45.43	0.535						0.4060
-43.82	0.644						0.4040
-47.59	0.417						0.4020
-70.25	C.031						0.4000
-46.61	0.467						0.3980
-43.57	0.663	*	*	*	*	*	0.3960
-45.79	0.514						0.3940
-60.65	0.092						0.3920
-48.36	0.382						0.3900
-43.61	0.660						0.3880
-44.52	0.594						0.3860
-53.34	C.215						0.3840
-51.00	0.282						0.3820
-43.95	0.625	*	*	*	*	*	0.3800
-43.64	C.657						0.3780
-49.54	0.334						0.3760
-55.39	0.170						0.3740
-44.59	0.589						0.3720
-62.08	0.702						0.3700
-47.04	0.445						0.3680
-65.81	0.051						0.3660
-45.60	0.525	*	*	*	*	*	0.3640
-42.78	0.726						0.3620
-45.27	0.545						0.3600
-62.73	0.073						0.3580
-47.07	0.443						0.3560
-42.73	0.730						0.3540
-43.97	0.633						0.3520
-54.07	0.198						0.3500
-49.20	0.347	*	*	*	*	*	0.3480
-42.93	0.714						0.3460
-43.03	0.705						0.3440
-49.86	0.221						0.3420
-52.45	0.238						0.3400
-43.37	0.678						0.3380
-42.37	C.761						0.3360
-47.14	0.439						0.3340
-58.31	0.121	*	*	*	*	*	0.3320
-44.10	0.674						0.3300
-41.95	0.799						0.3280
-45.70	C.550						0.3260
-70.81	0.033						0.3240
-45.18	0.557						0.3220











-50.71  
-45.84  
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C.292 |  
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REAL 99.9983

19.9588

39.9993

53

39.9998

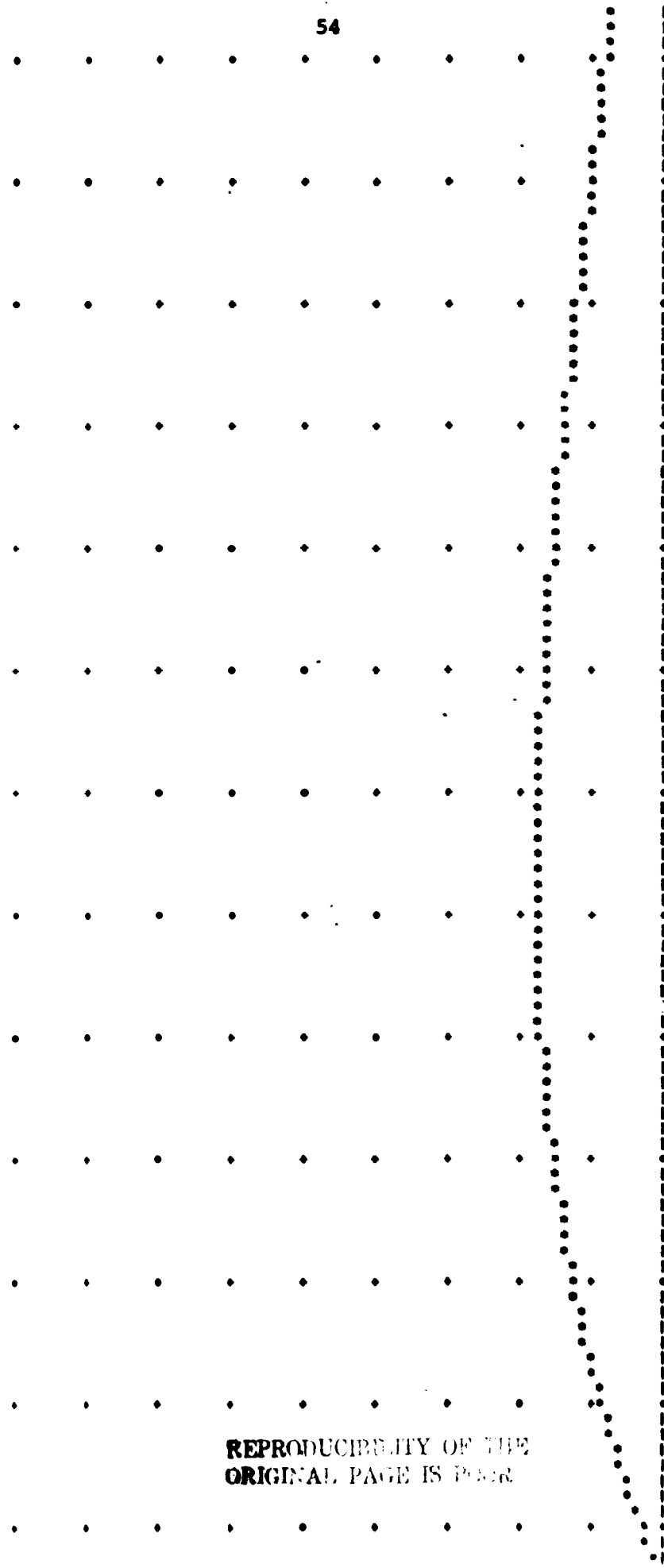
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Y-AXIS PROFILE PLCT ALONG 0.0 LONGITUDE.  
PATTERN NUMBER 100

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-33.67	0.021						0   3.9918
-34.42	0.019						0   3.9823
-35.26	0.017						0   3.9759
-36.19	0.015		•	•	•	•	0 • 3.9679
-37.26	0.014						0   3.9599
-38.50	0.012						0   3.9519
-39.97	0.010						0   3.9439
-41.76	0.003						0   3.9359
-44.06	0.006						0   3.9279
-47.24	0.004						0   3.9199
-52.42	0.002						0   3.9119
-67.60	0.000		•	•	•	•	0 • 3.9039
-56.02	0.002						0   3.8959
-48.86	0.004						0   3.8879
-44.95	0.006						0   3.8799
-42.23	0.009						0   3.8719
-40.15	0.010						0   3.8639
-38.65	0.012						0   3.8559
-37.02	0.014						0   3.8479
-35.78	0.016		•	•	•	•	0 • 3.8399
-34.68	0.018						0   3.8319
-33.70	0.021						0   3.8239
-32.81	0.023						0   3.8159
-31.99	0.025						0   3.8079
-31.24	0.027						0   3.7999
-30.54	0.030						0   3.7919
-29.84	0.032						0   3.7839
-29.28	0.034		•	•	•	•	0 • 3.7759
-28.71	0.037						0   3.7679
-28.16	0.039						0   3.7599
-27.65	0.041						0   3.7519
-27.16	0.044						0   3.7439
-26.70	0.046						0   3.7359
-26.25	0.049						0   3.7279
-25.83	0.051						0   3.7199
-25.42	0.054		•	•	•	•	0 • 3.7119
-25.03	0.056						0   3.7039
-24.65	0.059						0   3.6959
-24.24	0.061						0   3.6879
-23.84	0.064						0   3.6799

-23.41 0.066  
 -23.24 0.069  
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 -22.67 0.074  
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 -22.09 0.079  
 -21.82 0.081  
 -21.55 0.084  
 -21.25 0.086  
 -21.04 0.089  
 -20.80 0.091  
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 -20.33 0.096  
 -20.11 C.099  
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 -19.68 0.104  
 -19.48 0.106  
 -19.28 0.109  
 -19.09 0.111  
 -18.90 0.113  
 -18.72 0.116  
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 -17.45 0.134  
 -17.32 C.136  
 -17.18 C.138  
 -17.06 C.140  
 -16.93 C.142  
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 -16.11 C.156  
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 -15.88 0.161  
 -15.81 0.162  
 -15.75 C.163  
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 -15.34 C.170  
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 -15.51 C.168  
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 -18.16 C.124  
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 -19.92 C.101  
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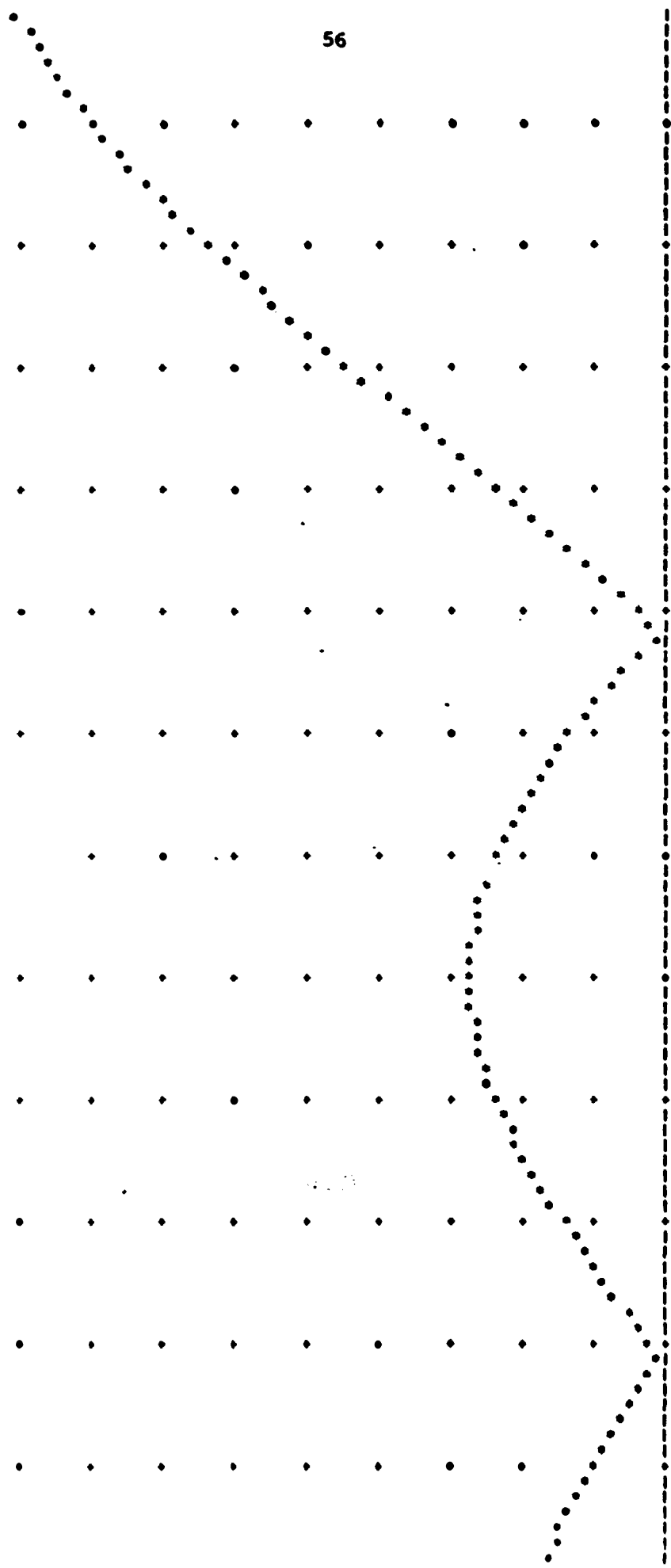


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REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR



-0.88	0.507
-1.71	0.470
-1.14	0.477
-1.29	0.462
-1.44	0.447
-1.61	0.431
-1.79	0.414
-1.97	0.397
-2.17	0.379
-2.38	0.360
-2.61	0.340
-2.85	0.320
-3.10	0.300
-3.37	0.278
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-3.95	0.234
-4.27	0.211
-4.61	0.188
-4.97	0.164
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-6.18	0.091
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-7.65	0.014
-8.21	0.289
-8.81	0.363
-9.45	0.437
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-13.67	0.807
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-21.64	1.182
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-11.95	3.541
-12.17	3.617
-12.42	3.693
-12.72	3.769
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-13.46	3.921
-13.90	3.997
-14.41	4.073
-14.98	4.149
-15.63	4.225
-16.36	4.301
-17.20	4.377
-18.15	4.453
-19.25	4.529
-20.55	4.605
-22.10	4.681
-24.01	4.757
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-44.90	5.061
-58.63	5.137
-80.52	5.213
-110.04	5.289
-154.82	5.365
-227.79	5.441
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-2214.41	5.973
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1.2480
1.2400
1.2320







EXAMPLE 2:  
DEMONSTRATION OF FOOTPRINT DEGRADATION DUE TO ANTENNA PANEL UNFOLDING ERRORS.

SYSTEM INFORMATION:  
FREQUENCY = 9.000 GHZ.  
YAW = 0.0 DEGREES  
TILT = 50.000 DEGREES  
TWIST = 0.0 DEGREES  
ALTITUDE = 200.000 KM.

ANTENNA PARAMETERS FOR SIMULATION NUMBER 101  
ELEMENT TYPE: HORIZONTAL DIPOLE  
NUMBER OF ELEMENTS (X,Y) = (504 , 12)  
INTERELEMENT SPACING (CM.): 2.2966 , 2.3550  
INTERELEMENT PHASE SHIFT (DEG.): 0.0 , 0.0

DEFORMATION DATA FOR SIMULATION 101:

-1.00	0.0	0.0	2.00
-1.00	0.0	0.0	2.00

ELECTRICAL DATA FOR SIMULATION 101:

	( 1, 1)	( 2, 1)	( 3, 1)	(
PMSX:	0.0	0.0	0.0	
PMSY:	0.0	0.0	0.0	
AMAC:	1.00	1.00	1.00	
APMS:	0.0	0.0	0.0	

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR

PRINT/PLCT INFORMATION:  
REQUESTED OUTPUT:  
PRINTER PROFILE  
PRINTER CCATCUR  
PLCT RESOLUTION: 151 X 151 POINTS

STARTX = -0.500  
STOPX = 0.500  
DELTAX = 0.007

STARTY = 0.0  
STOPY = 4.000  
DELTAY = 0.027

SUBARRAY DATA SUMMARY FOR PATTERN 101:

AREA	XCFNT	YCFNT	ZAVG	ALPHAX	ALPHAY	AMAC	APMS	PMSX	PMSY
1	-385.83	0.0	-0.50	0.148	0.0	1.0000	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0	0.0	1.0000	0.0	0.0	0.0
3	385.83	0.0	1.00	0.297	0.0	1.0000	0.0	0.0	0.0

PREDICTED BEAM CENTER:  
LATITUDE = 2.1545  
LONGITUDE = 0.0  
RANGE = 242.978

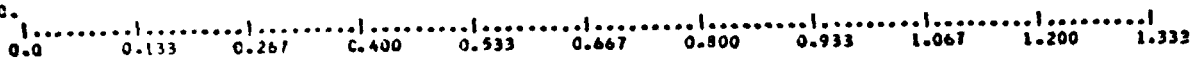
WEACING = 0.0

60

PLOT NORMALIZATION FACTOR = -46.325 00.



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0.2000.  
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0.3067.  
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0.4800.  
0.4867.  
0.4933.  
0.5000.



PRINTER CONTOUR PLOT FOR SIMULATION NUMBER 101

1.333 1.467 1.600 1.733 1.867 2.000 2.133 2.267 2.400 2.533 2.667

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR







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2.667 2.333 2.933 3.067 3.200 3.333 3.467 3.600 3.733 3.867 4.000



X-AXIS PROFILE PLOT ALONG 2.194 DEGREES LATITUDE.  
 PATTERN NUMBER 1C1

CE.	REAL	2.3224	1.8588	1.3951	G.9314	0.4678	0.0041	
-40.95	C.C07							C.5000
-32.78	O.C23							0.4920
-30.47	C.O30							0.4960
-30.25	O.O31							0.4940
-29.91	O.O32	•	•	•	•	•	•	0.4920
-28.59	O.O26							0.4900
-28.59	C.O37							0.4880
-29.39	O.O34							0.4860
-30.43	C.O20							0.4840
-30.17	O.O31							0.4820
-30.03	O.O22							0.4800
-32.24	O.O24							0.4780
-40.42	O.O10	•	•	•	•	•	•	0.4760
-39.46	O.O11							0.4740
-32.01	O.O25							0.4720
-29.94	O.O32							0.4700
-29.79	C.O32							0.4680
-29.44	O.O34							0.4660
-28.43	C.C38							0.4640
-28.13	C.C29							0.4620
-29.03	O.O35	•	•	•	•	•	•	0.4600
-29.92	O.O22							0.4580
-29.62	C.O33							0.4560
-29.65	O.O33							0.4540
-32.50	C.O24							0.4520
-42.65	C.O08							0.4500
-37.28	C.O14							0.4480
-31.05	C.C28							0.4460
-29.37	O.O34	•	•	•	•	•	•	0.4440
-29.31	O.O34							0.4420
-28.81	C.C34							0.4400
-27.60	O.O41							0.4380
-27.66	C.C41							0.4360
-28.72	C.O37							0.4340
-25.45	C.O21							0.4320
-26.02	O.O35							0.4300
-29.35	O.O34	•	•	•	•	•	•	0.4280
-32.72	O.O27							0.4260
-45.25	C.O05							0.4240
-35.03	O.O18							0.4220
-29.49	C.O32							0.4200
-29.78	O.O36							0.4180
-28.76	C.C36							0.4160
-27.07	C.C35							0.4140
-27.14	O.O44	•	•	•	•	•	•	0.4120
-27.25	C.O43							0.4100
-28.44	C.O38							0.4080
-28.91	O.O36							0.4060
-28.40	C.O38							0.4040
-29.22	C.O35							0.4020
-32.75	O.O21							0.4000
-46.45	O.O05							0.3980
-32.59	O.O23	•	•	•	•	•	•	0.3960
-28.92	O.O36							0.3940
-28.21	C.C39							0.3920
-28.15	O.O27							0.3900
-27.23	C.O44							0.3880
-26.48	C.O47							0.3860
-26.95	O.O45							0.3840
-28.14	C.O29							0.3820
-23.21	O.O39	•	•	•	•	•	•	0.3800
-27.82	O.O41							0.3780
-29.37	C.O34							0.3760
-35.01	O.O15							0.3740
-40.08	C.O10							0.3720
-30.53	O.O30							0.3700
-27.53	C.O40							0.3680
-27.64	C.O41							0.3660
-27.37	O.O43	•	•	•	•	•	•	0.3640
-26.31	C.O48							0.3620
-25.87	C.O51							0.3600
-26.70	O.O46							0.3580
-27.72	C.O41							0.3560
-27.60	C.O43							0.3540
-27.26	O.O43							0.3520
-30.07	C.O32							0.3500
-40.03	O.O10	•	•	•	•	•	•	0.3480
-34.82	O.O14							0.3460
-28.64	O.O37							0.3440
-27.06	O.O44							0.3420
-27.03	O.O45							0.3400
-26.41	O.O43							0.3380
-25.38	C.O54							0.3360
-25.38	C.O54							0.3340
-25.40	O.O47	•	•	•	•	•	•	0.3320
-27.04	O.O44							0.3300
-26.70	O.O47							0.3280
-27.3	O.O47							0.3260
-31.55	O.O50							0.3240
-40.27	O.O10							0.3220
-30.11	O.O10							0.3200
-27.1	O.O49							0.3180





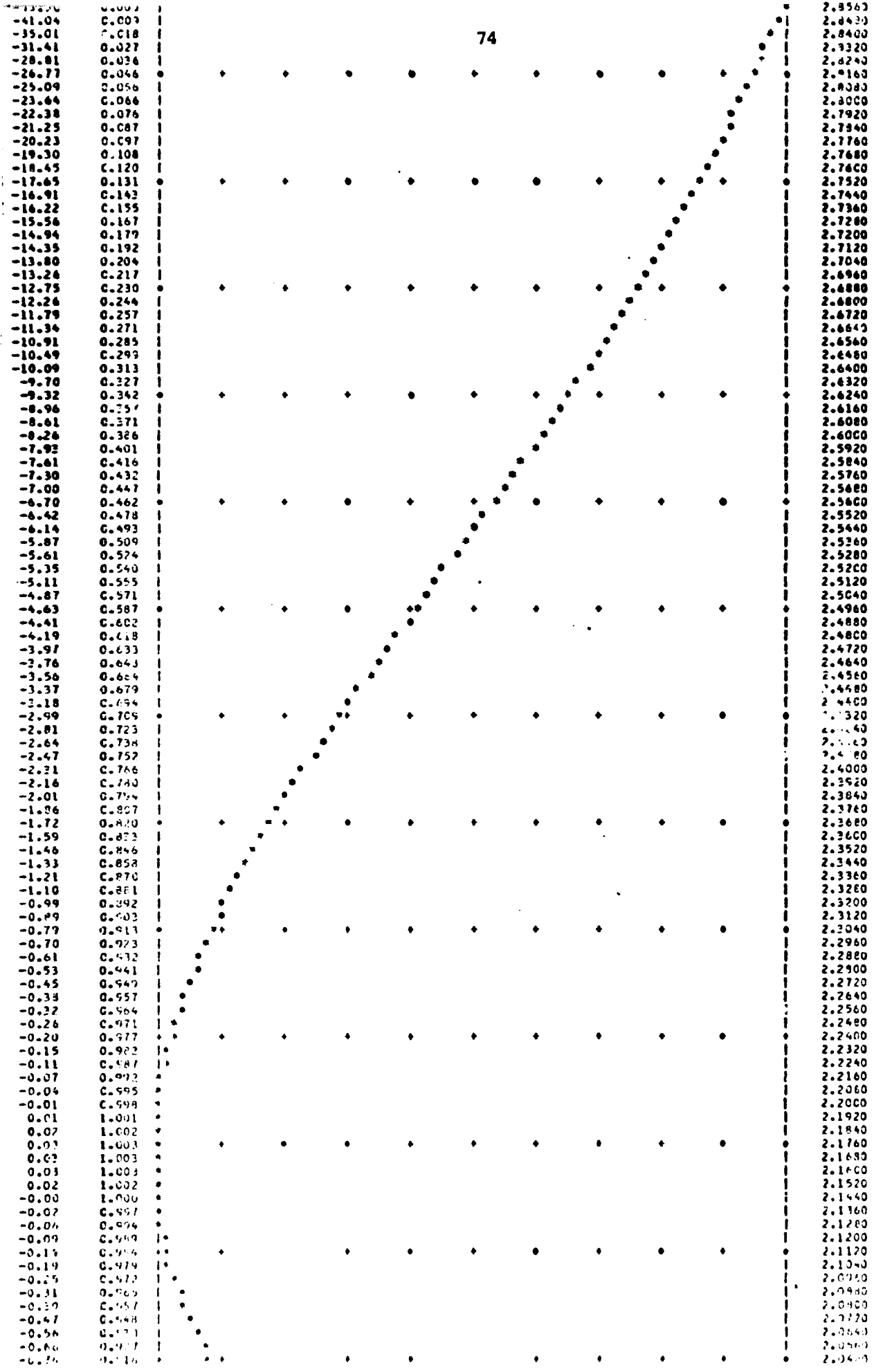




Y-AXIS PROFILE PLCT ALONG 0.0 LONGITUDE.  
 PATTERN NUMBER 101

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-37.00	0.014						3.9599
-38.24	0.012						3.9519
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-41.50	0.008						3.9359
-43.81	0.006						3.9279
-46.99	0.004						3.9199
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-67.35	0.000						3.9039
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-44.70	0.006						3.8799
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-24.82	0.057						3.7039
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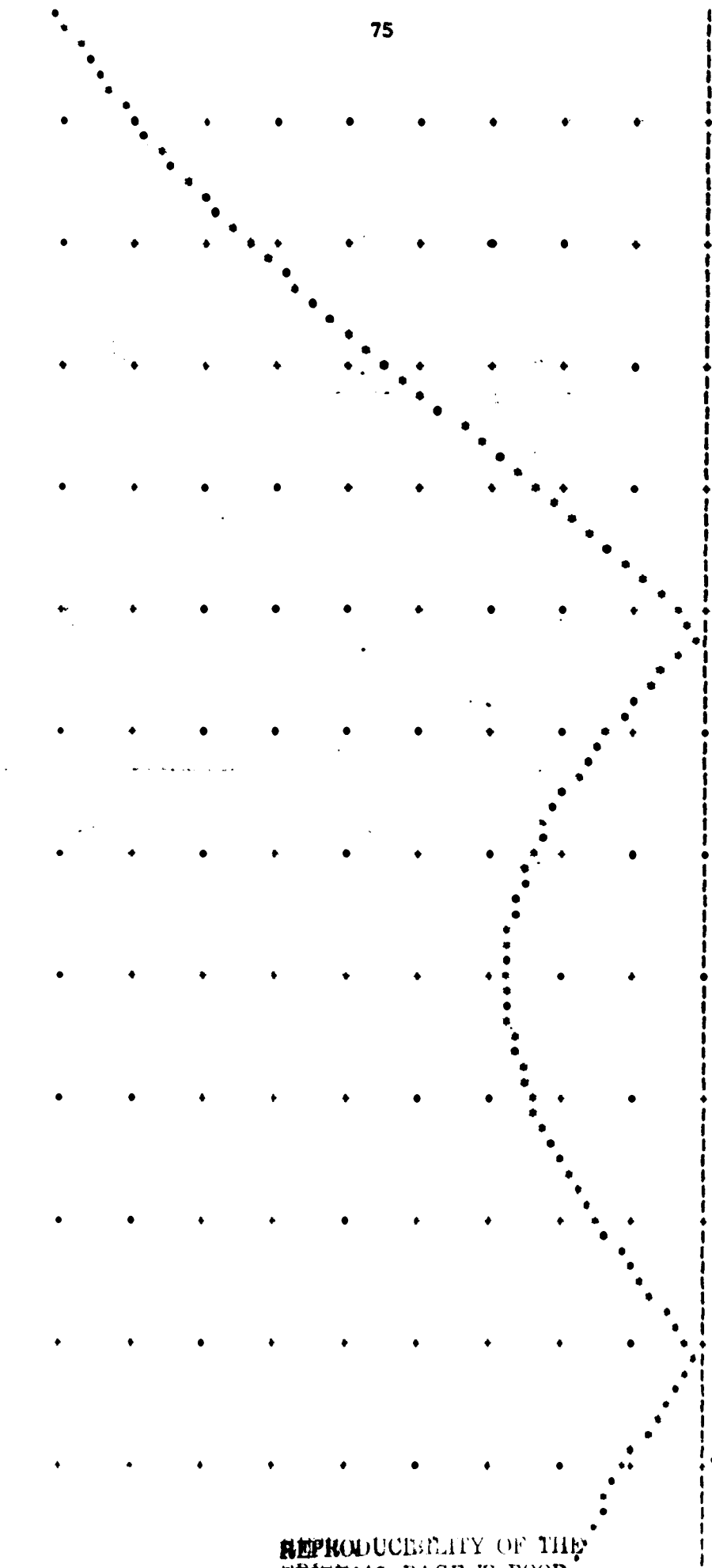
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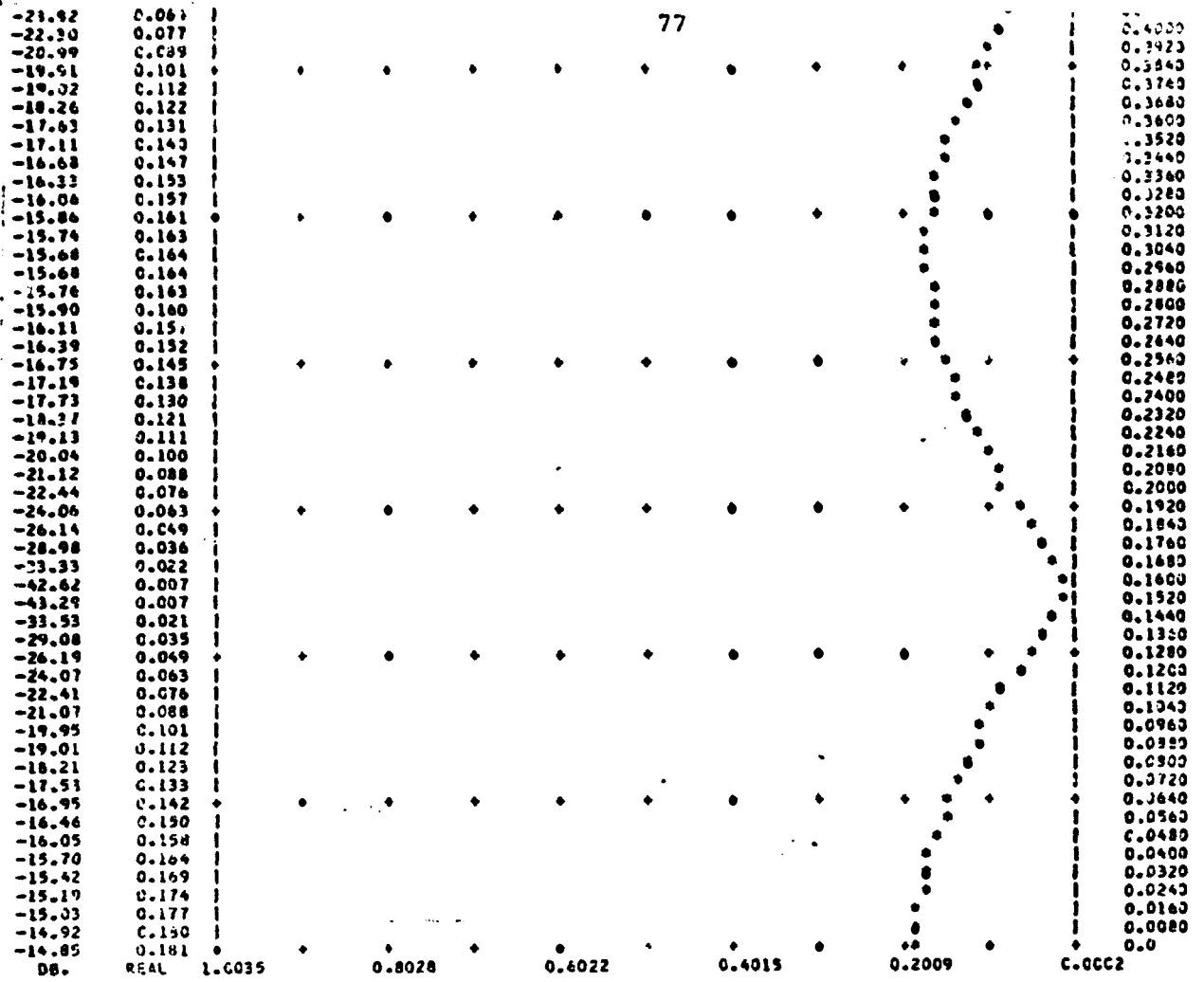
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-11.70	0.260
-12.61	0.234
-13.63	0.208
-14.77	0.183
-16.06	0.157
-17.57	0.132
-19.36	0.108
-21.59	0.093
-24.52	0.059
-28.85	0.036
-37.55	0.013
-40.94	0.009
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-15.44	0.169
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-19.04	0.112
-20.37	0.096
-21.88	0.081
-23.79	0.065
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-29.74	0.033
-35.63	0.017
-64.65	0.001
-36.38	0.015
-30.26	0.031
-26.77	0.046
-24.35	0.061
-22.51	0.075
-21.05	0.089
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-17.29	0.137
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-15.71	0.164



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1.2320

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR





EXAMP 3:  
 DEMONSTRATION OF THERMALLY INDUCED MECHANICAL DEFORMATIONS  
 (SPHERICAL) ON THE FULL X-BAND HORIZONTAL ARRAY.

SYSTEM INFORMATION:  
 FREQUENCY = 9.000 GHz.  
 YAW = 0.0 DEGREES  
 TILT = 50.000 DEGREES  
 TWIST = 0.0 DEGREES  
 ALTITUDE = 200.000 KM.

ANTENNA PARAMETERS FOR SIMULATION NUMBER 102  
 ELEMENT TYPE: HORIZONTAL DIPOLE  
 NUMBER OF ELEMENTS (X,Y) = (504 , 12)  
 INTERELEMENT SPACING (CM) = 2.2966 , 2.3550  
 INTERELEMENT PHASE SHIFT (DEG) = 0.0 , 0.0

DEFORMATION DATA FOR SIMULATION 102:

5.00	3.47	2.22	1.25	0.55	0.14	0.0	0.14	0.55	1.25
2.22	3.47	5.00							
5.00	3.47	2.22	1.25	0.55	0.14	0.0	0.14	0.55	1.25
2.22	3.47	5.00							

REPRODUCIBILITY OF DATA  
 ORIGINAL PAGE IS FOR...

ELECTRICAL DATA FOR SIMULATION 102:

	( 1, 1)	( 2, 1)	( 3, 1)	( 4, 1)	( 5, 1)	( 6, 1)	( 7, 1)	( 8, 1)	( 9, 1)	(10, 1)
PHSX:	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
PHSY:	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
ANAG:	1.00	1.00	1.00	1.00	1.00	1.00	1.00	1.00	1.00	1.00
APHS:	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

PRINT/PLT INFORMATION:  
 REQUESTED OUTPUT:  
 PRINTER PROFILE  
 PRINTER CONTROL  
 PLOT RESOLUTION: 151 X 151 POINTS

STARTX = -0.500  
 STOPX = 0.500  
 DELTAX = 0.002  
 STARTY = 0.0  
 STOPY = 4.000  
 DELTAY = 0.002

SUMMARY DATA SUMMARY FOR PATTERN 102:

AREA	AVG	YAW	TILT	TWIST	ALTITUDE	PHSX	PHSY	ANAG	APHS
1	0.0	0.0	50.0	0.0	200.0	0.0	0.0	1.000	0.0
2	0.0	0.0	50.0	0.0	200.0	0.0	0.0	1.000	0.0

3	-337.60	0.0	1.74	-0.577	0.0	1.0000	0.0	0.0	0.0
4	-241.14	0.0	0.90	-0.413	0.000	1.0000	0.0	0.0	0.0
5	-144.65	0.0	0.35	-0.247	0.0	1.0000	0.0	0.0	0.0
6	-48.23	0.0	0.07	-0.082	0.0	1.0000	0.0	0.0	0.0
7	48.23	0.0	0.07	0.082	0.0	1.0000	0.0	0.0	0.0
8	144.65	0.0	0.35	0.247	0.0	1.0000	0.0	0.0	0.0
9	241.14	0.0	0.90	0.413	0.000	1.0000	0.0	0.0	0.0
10	337.60	0.0	1.74	0.577	0.0	1.0000	0.0	0.0	0.0
11	434.04	0.0	2.05	0.762	0.0	1.0000	0.0	0.0	0.0
12	530.51	0.0	4.24	0.908	0.0	1.0000	0.0	0.0	0.0

PREDICTED BEAP CENTER:  
 LATITUDE = 2.1945  
 LONGITUDE = 0.0  
 RANGE = 213.978  
 BEACING = 0.0

PLGT NORMALIZATION FACTOR = -51.369 DB.



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 0.1267.0 C  
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 0.1533.  
 C.1600.  
 0.1667.  
 C.1733.  
 0.1800.  
 0.1867.  
 0.1933.  
 0.2000.  
 0.2067.  
 0.2133.  
 0.2200.  
 0.2267.  
 0.2333.  
 0.2400.  
 0.2467.  
 0.2533.  
 C.2600.  
 0.2667.  
 0.2733.  
 C.2800.  
 0.2867.  
 0.2933.  
 C.3000.  
 0.3067.  
 C.3133.  
 C.3200.  
 0.3267.  
 0.3333.  
 C.3400.  
 0.3467.  
 0.3533.  
 0.3600.  
 0.3667.  
 0.3733.  
 0.3800.  
 C.3867.  
 0.3933.  
 0.4000.  
 0.4067.  
 0.4133.  
 C.4200.  
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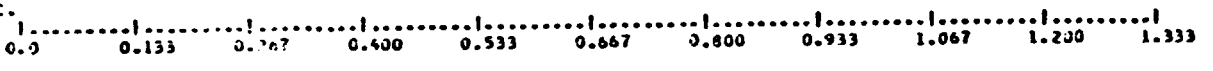
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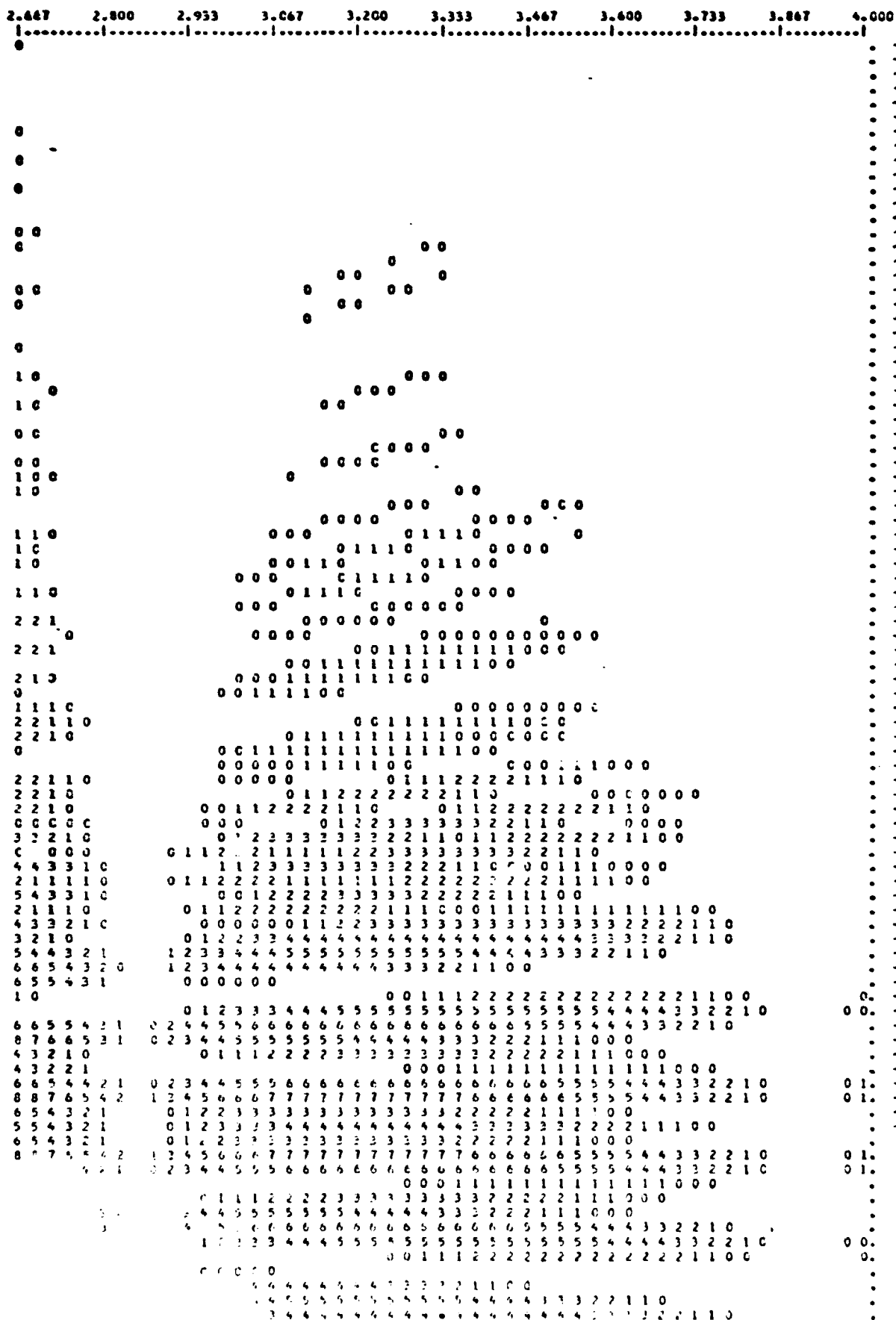








PRINTER CONTOUR PLOT FOR SIMULATION NUMBER 102





λ-AXIS PROFILE PLOT ALONG 2.194 DEGREES LATITUDE.  
PATTERN NUMBER 102

DB.	REAL	2.9574	2.3671	1.7768	1.1865	0.5943	0.0060
-19.08	0.111						0.5000
-16.90	0.143						0.4980
-17.52	0.133						0.4960
-21.07	0.088						0.4940
-29.89	0.032	*	*	*	*	*	0.4920
-24.43	0.048						0.4900
-20.90	0.090						0.4880
-18.99	0.112						0.4860
-18.93	0.113						0.4840
-20.45	0.095						0.4820
-24.28	0.061						0.4800
-35.28	0.017						0.4780
-29.79	0.032	*	*	*	*	*	0.4760
-23.03	0.071						0.4740
-20.60	0.093						0.4720
-19.73	0.103						0.4700
-19.07	0.111						0.4680
-18.07	0.125						0.4660
-17.36	0.136						0.4640
-17.37	0.135						0.4620
-14.52	0.119	*	*	*	*	*	0.4600
-20.82	0.091						0.4580
-24.34	0.061						0.4560
-29.64	0.033						0.4540
-41.03	0.009						0.4520
-35.48	0.017						0.4500
-27.20	0.044						0.4480
-22.57	0.074						0.4460
-19.53	0.106	*	*	*	*	*	0.4440
-17.75	0.120						0.4420
-17.16	0.139						0.4400
-17.63	0.131						0.4380
-18.67	0.117						0.4360
-19.34	0.108						0.4340
-19.56	0.105						0.4320
-20.62	0.093						0.4300
-24.29	0.061	*	*	*	*	*	0.4280
-36.78	0.014						0.4260
-26.57	0.047						0.4240
-20.41	0.095						0.4220
-17.08	0.123						0.4200
-17.25	0.137						0.4180
-16.42	0.120						0.4160
-22.55	0.074						0.4140
-44.05	0.006	*	*	*	*	*	0.4120
-22.00	0.179						0.4100
-16.60	0.148						0.4080
-24.94	0.179						0.4060
-15.99	0.159						0.4040
-20.96	0.090						0.4020
-34.11	0.020						0.4000
-19.40	0.107						0.3980
-15.43	0.168	*	*	*	*	*	0.3960
-14.97	0.178						0.3940
-17.17	0.139						0.3920
-23.28	0.069						0.3900
-29.67	0.037						0.3880
-20.68	0.093						0.3860
-17.61	0.132						0.3840
-16.83	0.144						0.3820
-17.66	0.131	*	*	*	*	*	0.3800
-20.50	0.094						0.3780
-27.96	0.040						0.3760
-30.99	0.023						0.3740
-21.60	0.083						0.3720
-19.41	0.120						0.3700
-17.30	0.136						0.3680
-16.77	0.145						0.3660
-15.99	0.159	*	*	*	*	*	0.3640
-15.24	0.173						0.3620
-15.24	0.173						0.3600
-14.27	0.154						0.3580
-18.44	0.120						0.3560
-21.82	0.031						0.3540
-26.05	0.045						0.3520
-37.42	0.013						0.3500
-23.14	0.022	*	*	*	*	*	0.3480
-24.84	0.057						0.3460
-20.28	0.097						0.3440
-17.31	0.136						0.3420
-15.60	0.166						0.3400
-15.07	0.176						0.3380
-15.57	0.166						0.3360
-16.50	0.150						0.3340
-14.90	0.143	*	*	*	*	*	0.3320
-16.99	0.141						0.3300
-18.24	0.122						0.3280
-22.54	0.075						0.3260
-14.51	0.077						0.3240
-17.07	0.074						0.3220
-17.10	0.140						0.3200
-14.11	0.176						0.3180









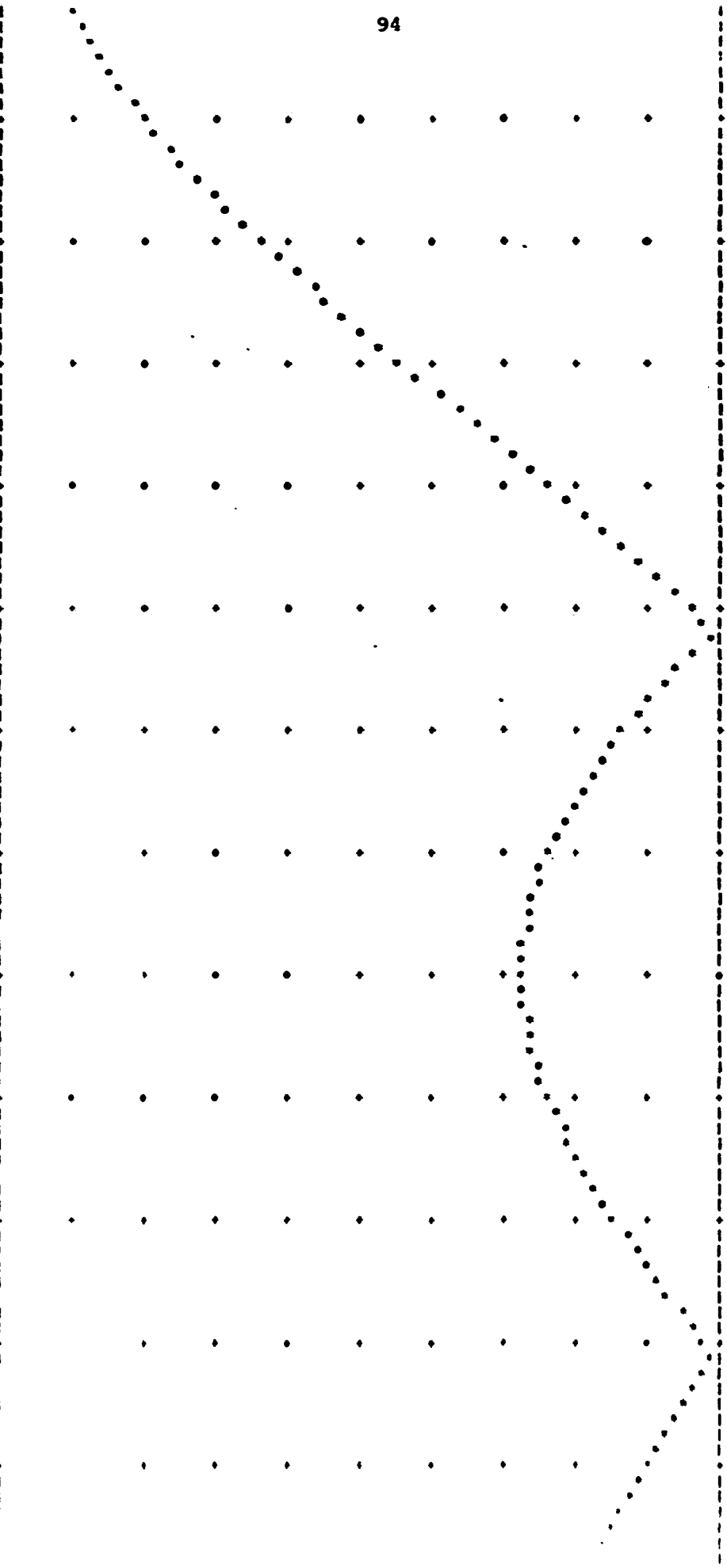






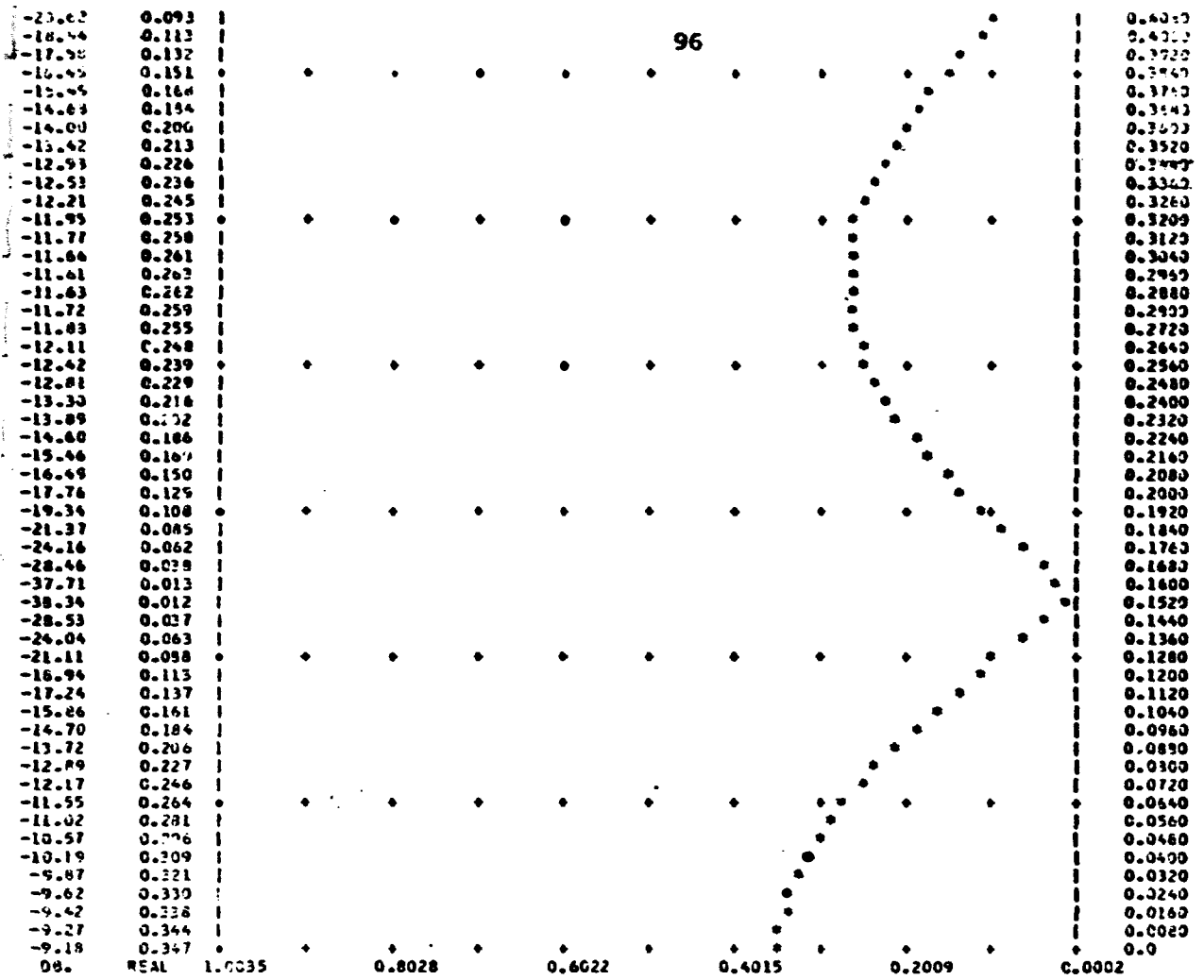


-0.89 C.903  
 -1.71 C.850  
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 -1.73 C.882  
 -1.45 C.867  
 -1.61 C.821  
 -1.79 C.814  
 -1.94 C.777  
 -2.16 C.778  
 -2.39 C.760  
 -2.61 C.740  
 -2.85 C.720  
 -3.11 C.699  
 -3.37 C.678  
 -3.66 C.656  
 -3.96 C.634  
 -4.28 C.611  
 -4.62 C.588  
 -4.98 C.564  
 -5.36 C.540  
 -5.76 C.515  
 -6.19 C.490  
 -6.65 C.465  
 -7.14 C.440  
 -7.66 C.414  
 -8.22 C.388  
 -8.81 C.362  
 -9.46 C.334  
 -10.16 C.310  
 -10.92 C.284  
 -11.75 C.259  
 -12.66 C.233  
 -13.68 C.207  
 -14.82 C.182  
 -16.12 C.155  
 -17.63 C.131  
 -19.42 C.107  
 -21.65 C.083  
 -24.58 C.059  
 -28.92 C.035  
 -37.62 C.010  
 -41.01 C.009  
 -30.37 C.020  
 -25.84 C.051  
 -22.97 C.071  
 -20.89 C.090  
 -19.27 C.105  
 -17.98 C.125  
 -16.90 C.143  
 -15.99 C.159  
 -15.22 C.175  
 -14.55 C.187  
 -13.98 C.200  
 -13.49 C.212  
 -13.06 C.222  
 -12.70 C.232  
 -12.38 C.240  
 -12.12 C.248  
 -11.91 C.254  
 -11.74 C.259  
 -11.61 C.263  
 -11.52 C.265  
 -11.47 C.267  
 -11.46 C.267  
 -11.48 C.267  
 -11.54 C.265  
 -11.63 C.262  
 -11.77 C.259  
 -11.94 C.253  
 -12.15 C.247  
 -12.40 C.240  
 -12.70 C.232  
 -13.04 C.223  
 -13.43 C.213  
 -13.87 C.202  
 -14.38 C.191  
 -14.94 C.179  
 -15.59 C.166  
 -16.32 C.153  
 -17.15 C.139  
 -18.10 C.124  
 -19.20 C.110  
 -20.49 C.095  
 -22.03 C.079  
 -23.94 C.064  
 -26.42 C.049  
 -29.40 C.032  
 -35.79 C.016  
 -64.81 C.001  
 -36.53 C.015  
 -30.42 C.020  
 -26.53 C.035  
 -24.50 C.060  
 -22.66 C.074  
 -21.70 C.087  
 -20.81 C.100  
 -19.00 C.112  
 -18.15 C.124  
 -17.44 C.136  
 -16.81 C.148  
 -16.21 C.160



2.0400  
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 1.2800  
 1.2720  
 1.2640  
 1.2560  
 1.2480  
 1.2400





NORMAL TERMINATION

PATTERN 103

SPACE SHUTTLE IMAGING RADAR ANTENNA SIMULATION PROGRAM

25 OCTOBER 1976

EXAMPLE 4:  
DEMONSTRATION OF SYSTEMATIC ELECTRICAL ERRORS IN THE FULL X-BAND  
HORIZONTAL ARRAY.

SYSTEM INFORMATION:  
FREQUENCY = 9.000 GHZ.  
YAW = 0.0 DEGREES  
TILT = 50.000 DEGREES  
TWIST = 0.0 DEGREES  
ALTITUDE = 200.000 KM.

ANTENNA PARAMETERS FOR SIMULATION NUMBER 103  
ELEMENT TYPE: HORIZONTAL DIPOLE  
NUMBER OF ELEMENTS (X,Y) = 1504 , 12  
INTERELEMENT SPACING (CM.): 2.2966 , 2.3550  
INTERELEMENT PHASE SHIFT (DEG.): 0.0 , 0.0

## DEFORMATION DATA FOR SIMULATION 103:

0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0

## ELECTRICAL DATA FOR SIMULATION 103:

	( 1, 2)	( 2, 2)	( 3, 2)	( 4, 2)	( 5, 2)	( 6, 2)	(
PHSX:	0.0	0.0	0.0	0.0	0.0	0.0	0.0
PHSY:	0.0	0.0	0.0	0.0	0.0	0.0	0.0
AMAG:	1.00	1.00	1.00	1.00	1.00	1.00	1.00
APHS:	15.00	10.00	5.00	0.0	0.0	0.0	0.0

	( 1, 1)	( 2, 1)	( 3, 1)	( 4, 1)	( 5, 1)	( 6, 1)	(
PHSX:	0.0	0.0	0.0	0.0	0.0	0.0	0.0
PHSY:	0.0	0.0	0.0	0.0	0.0	0.0	0.0
AMAG:	1.00	1.00	1.00	1.00	1.00	1.00	1.00
APHS:	0.0	0.0	0.0	-5.00	-10.00	-15.00	

PRINT/FILE INFORMATION:  
REQUESTED PLOTFILE:  
PRINTED PLOTFILE:  
PRINTED COUNTER:  
PLOT RESOLUTION: 151 X 151 POINTS

STAIRX = -0.500  
STAIRY = 0.500  
DELTA X = 0.007

STAIRY = 0.0  
STAIRX = 0.007  
DELTA Y = 0.007

REPRODUCIBILITY OF THE  
ORIGINAL PAGE IS POOR

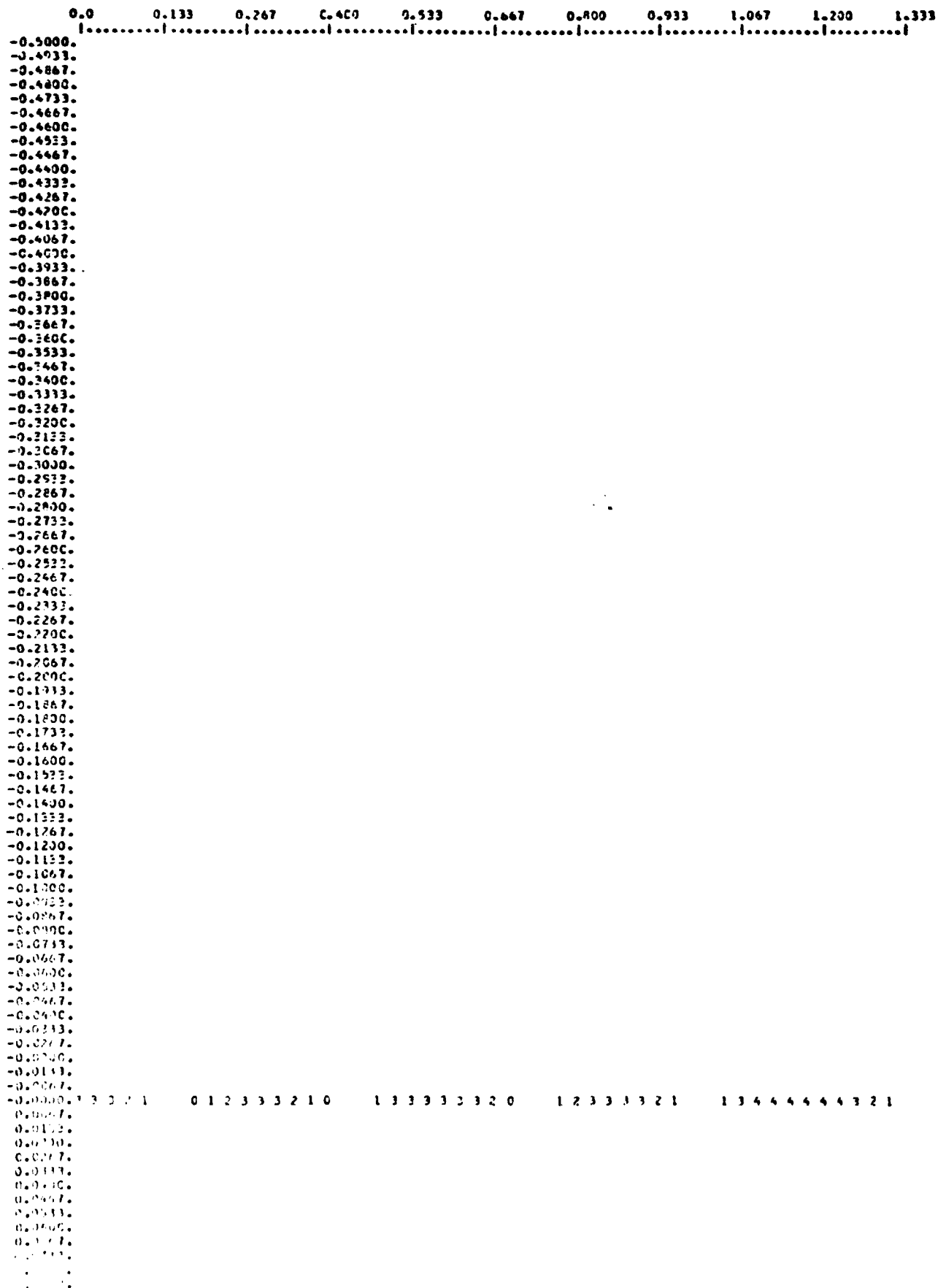
AREA	XCFNT	YCFNT	ZAVG	ALPHXK	ALPHBY	APST	APNS	PMSX	PMSY
1	-482.29	-7.06	0.0	0.0	0.0	1.0000	0.0	0.0	0.0
2	-289.37	-7.06	0.0	0.0	0.0	1.0000	0.0	0.0	0.0
3	-96.46	-7.06	0.0	0.0	0.0	1.0000	0.0	0.0	0.0
4	56.46	-7.06	0.0	0.0	0.0	1.0000	-5.00	0.0	0.0
5	289.37	-7.06	0.0	0.0	0.0	1.0000	-10.00	0.0	0.0
6	482.29	-7.06	0.0	0.0	0.0	1.0000	-15.00	0.0	0.0
7	-482.29	7.06	0.0	0.0	0.0	1.0000	15.00	0.0	0.0
8	-289.37	7.06	0.0	0.0	0.0	1.0000	10.00	0.0	0.0
9	-96.46	7.06	0.0	0.0	0.0	1.0000	5.00	0.0	0.0
10	96.46	7.06	0.0	0.0	0.0	1.0000	0.0	0.0	0.0
11	289.37	7.06	0.0	0.0	0.0	1.0000	0.0	0.0	0.0
12	482.29	7.06	0.0	0.0	0.0	1.0000	0.0	0.0	0.0

PREDICTED BEAP CENTER:  
 LATITUDE = 2.1945  
 LONGITUDE = 0.0  
 RANGE = 243.578  
 HEADING = 0.0

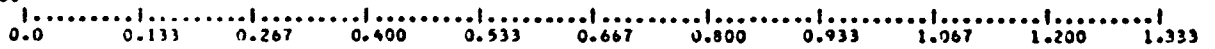
PLCT NORMALIZATION FACTOR = -35.088 DB.



PRINTED CONTOUR PLOT FOR SIMULATION NUMBER 103

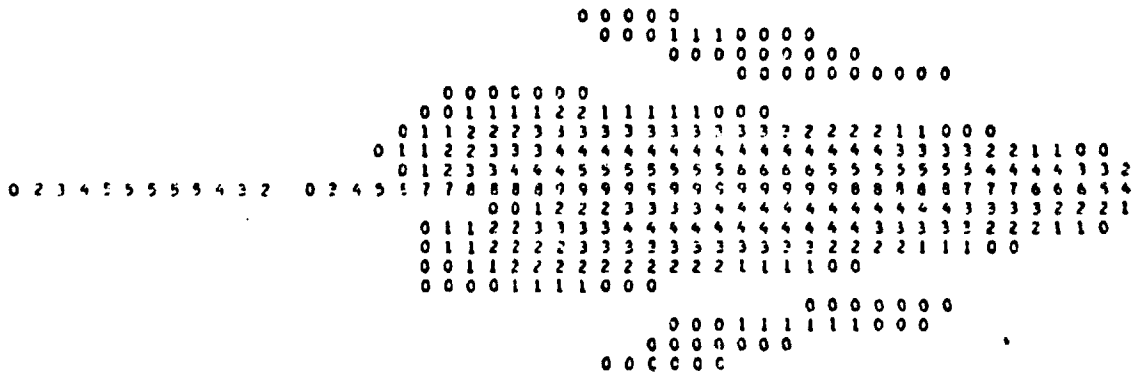


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0.1267.  
0.1333.  
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0.1467.  
0.1533.  
0.1600.  
0.1667.  
0.1733.  
0.1800.  
0.1867.  
0.1933.  
0.2000.  
0.2067.  
0.2133.  
0.2200.  
0.2267.  
0.2333.  
0.2400.  
0.2467.  
0.2533.  
0.2600.  
0.2667.  
0.2733.  
0.2800.  
0.2867.  
0.2933.  
0.3000.  
0.3067.  
0.3133.  
0.3200.  
0.3267.  
0.3333.  
0.3400.  
0.3467.  
0.3533.  
0.3600.  
0.3667.  
0.3733.  
0.3800.  
0.3867.  
0.3933.  
0.4000.  
0.4067.  
0.4133.  
0.4200.  
0.4267.  
0.4333.  
0.4400.  
0.4467.  
0.4533.  
0.4600.  
0.4667.  
0.4733.  
0.4800.  
0.4867.  
0.4933.  
0.5000.



PRINTER CONTOUR PLOT FOR SIMULATION NUMBER 103

1.333    1.467    1.600    1.733    1.867    2.000    2.133    2.267    2.400    2.533    2.667  
|.....|.....|.....|.....|.....|.....|.....|.....|.....|.....|

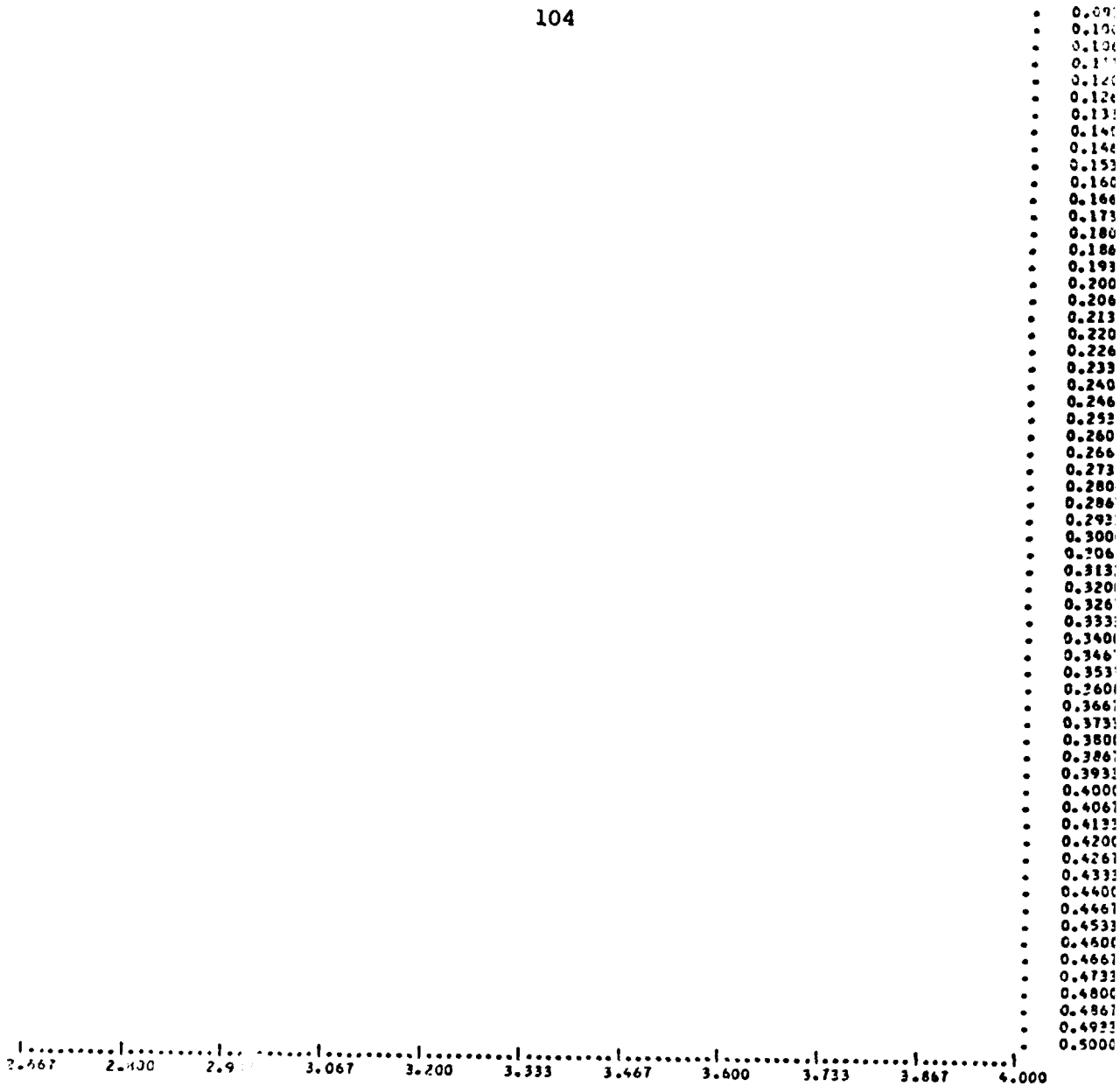


.....|.....|.....|.....|.....|.....|.....|.....|.....|.....|.....|  
 1.333 1.467 1.600 1.733 1.867 2.000 2.133 2.267 2.400 2.533 2.667

CONTOUR LEVEL KEY

0: -0.2850000E 02 TO -0.2549998E 02	4: -0.1650000E 02 TO -0.1349999E 02	8: -0.4500000E 01 TO -0.1499999E 01
1: -0.2550000E 02 TO -0.2249998E 02	5: -0.1350000E 02 TO -0.1049999E 02	9: -0.1500000E 01 TO 0.1500000E 01
2: -0.2250000E 02 TO -0.1949998E 02	6: -0.1050000E 02 TO -0.7499999E 01	: -0.1000000E 01 TO -0.2950000E 02
3: -0.1950000E 02 TO -0.1649998E 02	7: -0.7500000E 01 TO -0.4499999E 01	+: 0.1500000E 01 TO 0.1000000E 01





Z-TYPE PROFILE PLOT ALONG 2.104 DEGREES LATITUDE.  
PATTERN NUMBER 103

SCALE FACTOR IS 10<sup>-2</sup>

CR.	FFM	79.9427	79.9560	79.9653	79.9826	79.9959	80.0392
-66.02	0.440						0.5000
-66.71	0.591						0.4900
-66.11	0.293						0.4960
-72.41	0.624						0.4940
-69.24	0.345	•	•	•	•	•	0.4920
-65.50	0.531						0.4900
-67.10	0.442						0.4880
-58.27	0.172						0.4860
-51.48	0.256						0.4840
-45.58	0.576						0.4820
-65.61	0.424						0.4800
-51.65	0.262						0.4780
-57.77	0.129	•	•	•	•	•	0.4760
-66.55	0.449						0.4740
-45.44	0.534						0.4720
-65.37	0.340						0.4700
-69.13	0.035						0.4680
-67.59	0.399						0.4660
-64.76	0.565						0.4640
-67.00	0.447						0.4620
-59.49	0.176	•	•	•	•	•	0.4600
-51.05	0.280						0.4580
-45.40	0.557						0.4560
-66.90	0.492						0.4540
-57.40	0.115						0.4520
-50.88	0.286						0.4500
-64.51	0.555						0.4480
-64.71	0.620						0.4460
-69.70	0.367	•	•	•	•	•	0.4440
-63.60	0.066						0.4420
-66.87	0.453						0.4400
-64.46	0.594						0.4380
-67.38	0.428						0.4360
-70.73	0.079						0.4340
-68.12	0.293						0.4320
-64.16	0.619						0.4300
-45.67	0.532	•	•	•	•	•	0.4280
-55.04	0.177						0.4260
-51.51	0.266						0.4240
-64.97	0.555						0.4220
-64.67	0.564						0.4200
-51.59	0.269						0.4180
-54.76	0.293						0.4160
-65.32	0.542						0.4140
-64.37	0.625	•	•	•	•	•	0.4120
-68.17	0.390						0.4100
-67.71	0.041						0.4080
-67.16	0.493						0.4060
-64.61	0.564						0.4040
-67.92	0.407						0.4020
-70.21	0.031						0.4000
-60.23	0.438						0.3980
-62.72	0.731	•	•	•	•	•	0.3960
-64.06	0.626						0.3940
-53.14	0.230						0.3920
-50.96	0.233						0.3900
-64.64	0.627						0.3880
61.95	0.694						0.3860
-50.41	0.302						0.3840
-54.62	0.199						0.3820
-64.36	0.604	•	•	•	•	•	0.3800
-63.68	0.702						0.3780
-67.15	0.439						0.3760
-66.24	0.040						0.3740
-65.52	0.505						0.3720
-63.73	0.689						0.3700
-65.98	0.533						0.3680
-67.79	0.041						0.3660
-66.50	0.457	•	•	•	•	•	0.3640
-67.91	0.716						0.3620
-64.20	0.609						0.3600
-54.50	0.188						0.3580
-64.84	0.221						0.3560
-63.63	0.547						0.3540
-64.60	0.517						0.3520
-54.70	0.129						0.3500
-64.97	0.156	•	•	•	•	•	0.3480
-67.20	0.768						0.3460
-61.76	0.830						0.3440
-66.61	0.667						0.3420
-57.92	0.206						0.3400
-64.30	0.066						0.3380
-64.19	0.777						0.3360
-64.60	0.206						0.3340
-70.25	0.217	•	•	•	•	•	0.3320
-64.01	0.697						0.3300
-61.7	0.040						0.3280
61.7	0.040						0.3260
61.7	0.040						0.3240
61.7	0.040						0.3220

REPRODUCTION OF  
ORIGINAL PANEL IS P







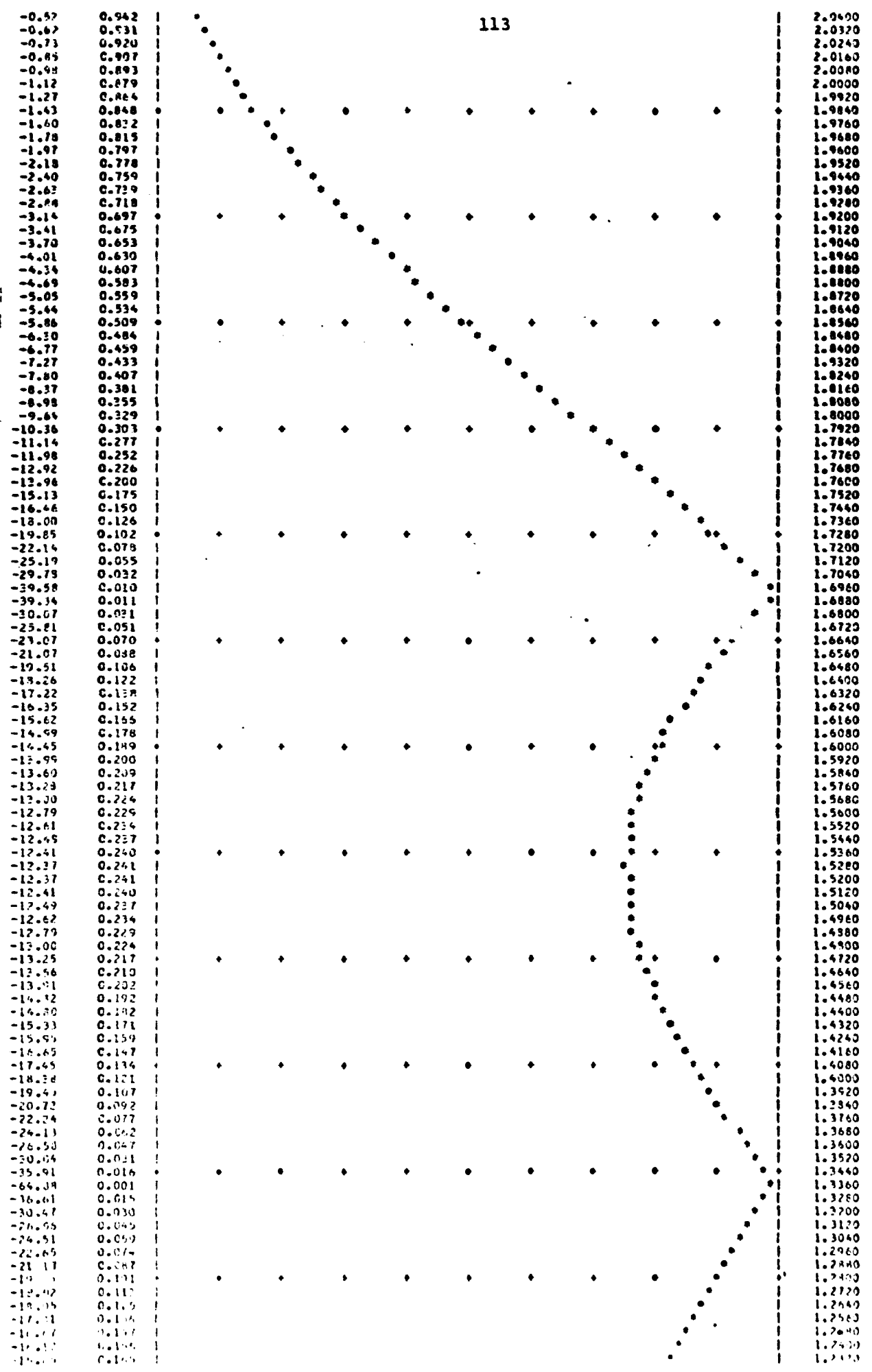












-0.57	0.942
-0.67	0.931
-0.73	0.920
-0.85	0.907
-0.98	0.893
-1.12	0.879
-1.27	0.864
-1.43	0.848
-1.60	0.832
-1.78	0.815
-1.97	0.797
-2.18	0.778
-2.40	0.759
-2.63	0.739
-2.88	0.718
-3.14	0.697
-3.41	0.675
-3.70	0.653
-4.01	0.630
-4.34	0.607
-4.69	0.583
-5.05	0.559
-5.44	0.534
-5.86	0.509
-6.30	0.484
-6.77	0.459
-7.27	0.433
-7.80	0.407
-8.37	0.381
-8.98	0.355
-9.64	0.329
-10.36	0.303
-11.14	0.277
-11.98	0.252
-12.92	0.226
-13.96	0.200
-15.13	0.175
-16.46	0.150
-18.00	0.126
-19.85	0.102
-22.14	0.078
-25.19	0.055
-29.79	0.032
-39.58	0.010
-39.34	0.011
-30.07	0.021
-25.81	0.051
-23.07	0.070
-21.07	0.088
-19.51	0.106
-18.26	0.122
-17.22	0.138
-16.35	0.152
-15.62	0.165
-14.99	0.178
-14.45	0.189
-13.99	0.200
-13.60	0.209
-13.29	0.217
-12.90	0.224
-12.79	0.229
-12.61	0.234
-12.45	0.237
-12.41	0.240
-12.37	0.241
-12.37	0.241
-12.41	0.240
-12.49	0.237
-12.62	0.234
-12.79	0.229
-13.00	0.224
-13.25	0.217
-13.56	0.210
-13.91	0.202
-14.32	0.192
-14.80	0.182
-15.33	0.171
-15.95	0.159
-16.65	0.147
-17.45	0.134
-18.38	0.121
-19.49	0.107
-20.72	0.092
-22.24	0.077
-24.13	0.062
-26.50	0.047
-30.04	0.031
-35.91	0.016
-64.04	0.001
-36.61	0.015
-30.47	0.030
-26.95	0.049
-24.51	0.069
-22.65	0.074
-21.17	0.087
-19.99	0.101
-18.92	0.117
-18.05	0.135
-17.31	0.156
-16.67	0.177
-16.13	0.199
-15.69	0.165

2.2400
2.0370
2.0240
2.0160
2.0080
2.0000
1.9920
1.9840
1.9760
1.9680
1.9600
1.9520
1.9440
1.9360
1.9280
1.9200
1.9120
1.9040
1.8960
1.8880
1.8800
1.8720
1.8640
1.8560
1.8480
1.8400
1.8320
1.8240
1.8160
1.8080
1.8000
1.7920
1.7840
1.7760
1.7680
1.7600
1.7520
1.7440
1.7360
1.7280
1.7200
1.7120
1.7040
1.6960
1.6880
1.6800
1.6720
1.6640
1.6560
1.6480
1.6400
1.6320
1.6240
1.6160
1.6080
1.6000
1.5920
1.5840
1.5760
1.5680
1.5600
1.5520
1.5440
1.5360
1.5280
1.5200
1.5120
1.5040
1.4960
1.4880
1.4800
1.4720
1.4640
1.4560
1.4480
1.4400
1.4320
1.4240
1.4160
1.4080
1.4000
1.3920
1.3840
1.3760
1.3680
1.3600
1.3520
1.3440
1.3360
1.3280
1.3200
1.3120
1.3040
1.2960
1.2880
1.2800
1.2720
1.2640
1.2560
1.2480
1.2400







PATTERN 104

SPACE SHUTTLE IMAGING RADAR ANTENNA SIMULATION PROGRAM

25 OCTOBER 1971

EXAMPLE 5:  
DEMONSTRATION OF THE EFFECT OF SHUTTLE ATTITUDE ERRORS ON FULL X-BAND  
HORIZONTAL ARRAY PERFORMANCE.

SYSTEM INFORMATION:  
FREQUENCY = 5.000 GHz.  
YAW = 2.000 DEGREES  
TILT = 52.000 DEGREES  
TWIST = 5.000 DEGREES  
ALTITUDE = 200.000 KM.

ANTENNA PARAMETERS FOR SIMULATION NUMBER 104  
ELEMENT TYPE: HORIZONTAL DIPOLE  
NUMBER OF ELEMENTS (X,Y) = (504 , 12)  
INTERELEMENT SPACING (CM.): 2.2966 , 2.3550  
INTERELEMENT PHASE SHIFT (DEG.): 0.0 , 0.0

GEOMETRIC DATA FOR SIMULATION 104:

0.0 0.0  
0.0 0.0

ELECTRICAL DATA FOR SIMULATION 104:

( 1, 1) ( 1  
PHSX: 0.0  
PHSY: 0.0  
AMAG: 1.00  
APHS: 0.0

PRINT/PLOT INFORMATION:  
REQUESTED OUTPUT:  
PRINTER PRCFILE  
PRINTER CCATCUR  
PLOT RESOLUTION: 151 X 151 POINTS

STARTX = -0.500  
STOPX = 0.500  
DELTAX = 0.007

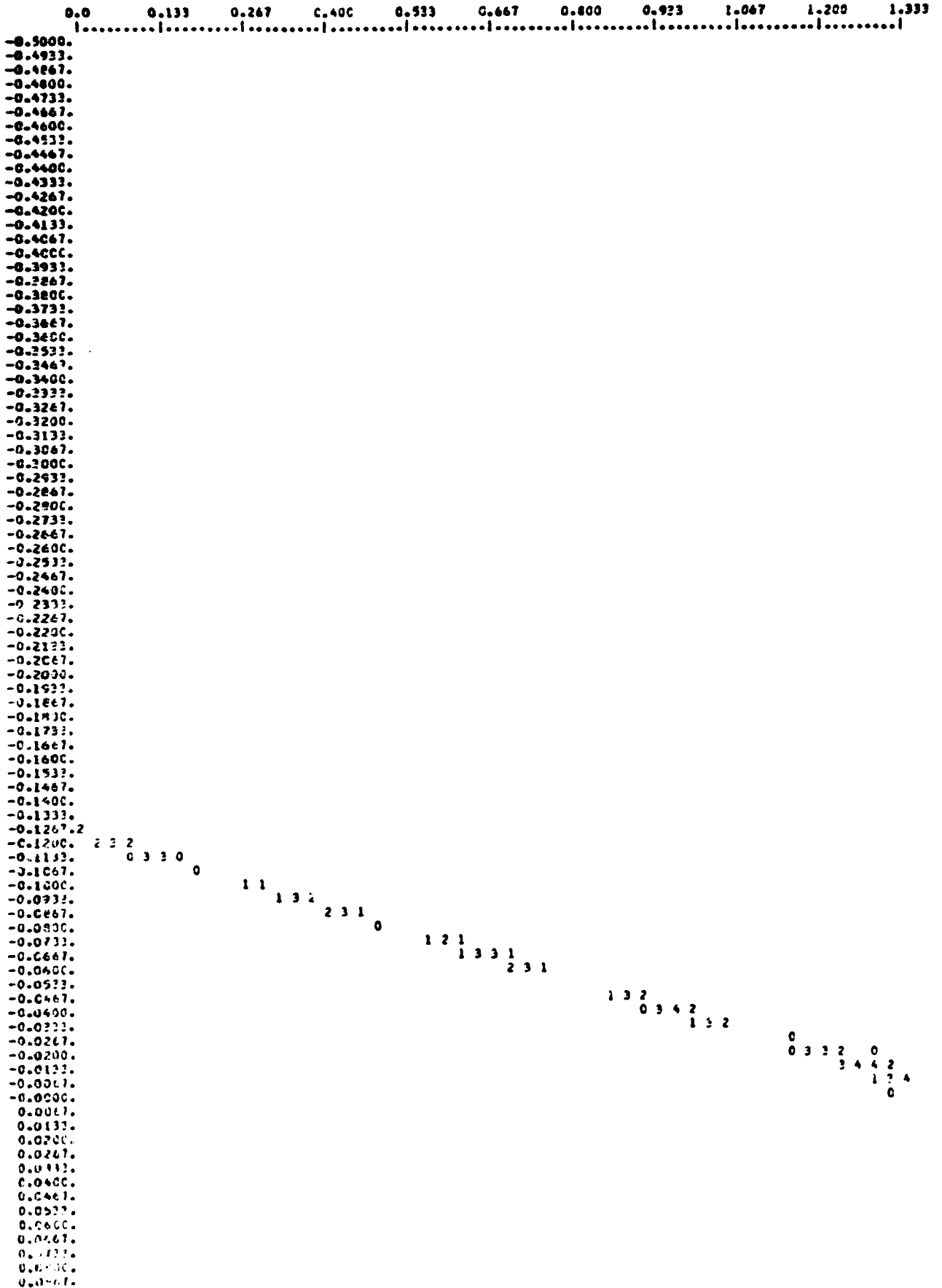
STARTY = 0.0  
STOPY = 4.000  
DELTAY = 0.027

AREA	XCENT	YCENT	ZAVG	ALPHAX	ALPHAY	AMAG	APHS	PHSX	PHSY
1	0.0	0.0	0.0	0.0	0.0	1.0000	0.0	0.0	0.0

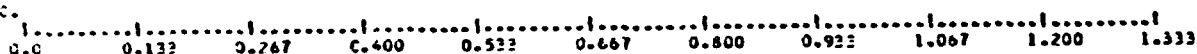
PREDICTED BEAM CENTER:  
LATITUDE = 2.3743  
LONGITUDE = 0.0026  
ALTITUDE = 2.2966  
TWIST = 5.0000

PLCT NORMALIZATION FACTOR = -35.418 DB.

PRINTED CONTOUR PLOT FOR SIMULATION NUMBER 104



0.0333.  
 0.1000.  
 0.1333.  
 0.1667.  
 0.2000.  
 0.2333.  
 0.2667.  
 0.3000.  
 0.3333.  
 0.3667.  
 0.4000.  
 0.4333.  
 0.4667.  
 0.5000.  
 0.5333.  
 0.5667.  
 0.6000.  
 0.6333.  
 0.6667.  
 0.7000.  
 0.7333.  
 0.7667.  
 0.8000.  
 0.8333.  
 0.8667.  
 0.9000.  
 0.9333.  
 0.9667.  
 1.0000.  
 1.0333.  
 1.0667.  
 1.1000.  
 1.1333.  
 1.1667.  
 1.2000.  
 1.2333.  
 1.2667.  
 1.3000.  
 1.3333.  
 1.3667.  
 1.4000.  
 1.4333.  
 1.4667.  
 1.5000.  
 1.5333.  
 1.5667.  
 1.6000.  
 1.6333.  
 1.6667.  
 1.7000.  
 1.7333.  
 1.7667.  
 1.8000.  
 1.8333.  
 1.8667.  
 1.9000.  
 1.9333.  
 1.9667.  
 2.0000.



**REPRODUCIBILITY OF THE  
 ORIGINAL PAGE**

PRINTED CONTOUR PLOT FOR SIMULATION NUMBER 104

1.333 1.467 1.600 1.733 1.867 2.000 2.133 2.267 2.400 2.533 2.667  
|-----|-----|-----|-----|-----|-----|-----|-----|-----|-----|

4 2  
C 2 3  
0 2 3 1 1 0  
2 4 5 4 1 1  
C 4 5 5 4 1 0  
1 1 4 5 4 3  
0 1 1 3 3 1  
0  
1 3 3 0 0 2 2 1 2 1 1 2 0 1 1 0 0 1 0 C  
1 4 5 6 5 2 1 3 2 2 3 2 1 2 1 0 1 1 0 0 0 C  
0 4 6 7 7 6 3 2 4 3 2 3 2 1 1 1 1 1 0 0  
C 2 0 3 6 8 8 6 3 3 4 4 2 3 2 1 1 0 2 1 0 1  
0 2 3 0 4 7 8 5 2 7 3 2 5 4 0 1 3 2 1 1 0  
1 0 3 4 5 4 4 4 4 7 3 1 4 4 0 1 2 0 1 1 0  
C 1 2 0 4 4 6 4 7 4 7 2 7 4 4 0 2 2 0 1 1 0  
1 0 2 5 4 2 7 5 4 6 2 1 4 3 0 1









X-AXIS PROFILE PLOT ALONG 2.364 DEGREES LATITUDE.  
PATTERN NUMBER 104

SCALE FACTOR IS 10<sup>00</sup>-2

DB.	PEAL	99.2487	79.3996	55.5506	39.7015	19.8525	0.0035
-45.30	C.538						0.5000
-54.79	0.182						0.4980
-51.62	C.267						0.4963
-46.55	0.589						0.4940
-43.83	0.643	*	*	*	*	*	0.4920
-48.03	C.257						0.4900
-48.28	0.619						0.4880
-46.74	C.460						0.4860
-43.47	C.670						0.4840
-44.91	0.568						0.4820
-54.08	0.194						0.4800
-51.38	C.270						0.4780
-44.22	0.615	*	*	*	*	*	0.4760
-43.42	C.675						0.4740
-47.55	0.417						0.4720
-47.98	C.040						0.4700
-46.32	C.493						0.4680
-43.05	0.704						0.4660
-44.52	C.594						0.4640
-53.85	0.203						0.4620
-50.79	0.789	*	*	*	*	*	0.4600
-43.75	C.645						0.4580
-43.01	0.707						0.4560
-47.32	C.430						0.4540
-45.48	C.055						0.4520
-45.73	0.517						0.4500
-42.60	0.741						0.4480
-44.21	C.616						0.4460
-54.13	0.197	*	*	*	*	*	0.4440
-45.87	C.321						0.4420
-43.20	C.492						0.4400
-42.83	0.758						0.4380
-47.25	0.434						0.4360
-41.86	0.001						0.4340
-44.95	0.543						0.4320
-42.13	C.783						0.4300
-43.59	0.637	*	*	*	*	*	0.4280
-55.00	C.178						0.4260
-48.71	C.367						0.4240
-42.58	C.743						0.4220
-42.20	C.749						0.4200
-47.39	0.427						0.4180
-58.19	C.123						0.4160
-44.13	C.622						0.4140
-41.63	0.828	*	*	*	*	*	0.4120
-42.87	0.441						0.4100
-56.65	0.144						0.4080
-47.36	C.423						0.4060
-41.90	0.402						0.4040
-41.99	0.756						0.4020
-47.81	C.407						0.4000
-54.73	C.132						0.3980
-43.18	0.697	*	*	*	*	*	0.3960
-41.15	C.874						0.3940
-43.88	C.640						0.3920
-60.70	0.002						0.3900
-45.85	0.509						0.3880
-41.20	0.871						0.3860
-41.72	C.619						0.3840
-48.61	C.371						0.3820
-51.64	0.262	*	*	*	*	*	0.3800
-42.18	0.778						0.3780
-40.68	C.925						0.3760
-44.04	0.527						0.3740
-72.70	0.021						0.3720
-44.39	0.604						0.3700
-40.45	0.845						0.3680
-41.55	0.833						0.3660
-50.01	0.116	*	*	*	*	*	0.3640
-48.83	C.182						0.3620
-41.16	0.745						0.3600
-40.26	0.970						0.3580
-44.51	0.565						0.3560
-62.62	0.074						0.3540
-42.88	0.718						0.3520
-39.81	1.003						0.3500
-41.58	0.824	*	*	*	*	*	0.3480
-52.53	0.236						0.3460
-46.27	C.495						0.3440
-40.15	C.983						0.3420
-39.52	1.006						0.3400
-45.31	0.643						0.3380
-54.19	0.195						0.3360
-41.41	C.100						0.3340
-39.17	1.101	*	*	*	*	*	0.3320
-41.75	0.810						0.3300
-57.72	0.130						0.3280
-42.08	0.682						0.3260
-39.13	1.007						0.3240
-39.70	1.005						0.3220









129  
 REAL 79.8408 79.8408 55.8806 39.9204 19.9603 0.0001  
 \*| -0.4960  
 \*| -0.4920  
 \* -0.5000  
 0.0035

Y-AXIS PROFILE PLCT ALONG 0.083LCNGITUDE.  
 PATTERN NUMBER 104

SCALE FACTOR IS 10\*\*2

DB	PEAK	99.8009	79.8408	55.8806	39.9204	19.9603	0.0001
-56.81	0.144						3.9958
-55.40	0.170						3.9918
-54.21	0.195						3.9838
-53.22	0.213						3.9759
-52.39	0.240	+	•	•	•	•	3.9679
-51.69	0.260						3.9599
-51.12	0.276						3.9519
-50.67	0.293						3.9439
-50.33	0.304						3.9359
-50.11	0.312						3.9279
-49.99	0.317						3.9199
-49.99	0.317						3.9119
-50.19	0.312	•	•	•	•	•	3.9039
-50.55	0.304						3.8959
-50.74	0.290						3.8879
-51.30	0.272						3.8799
-52.05	0.250						3.8719
-53.05	0.222						3.8639
-54.39	0.191						3.8559
-56.17	0.155						3.8479
-59.79	0.116	•	•	•	•	•	3.8399
-62.79	0.075						3.8319
-71.11	0.024						3.8239
-74.15	0.020						3.8159
-83.79	0.009						3.8079
-98.51	0.011						3.7999
-95.45	0.109						3.7919
-93.77	0.219						3.7839
-91.50	0.367	•	•	•	•	•	3.7759
-90.12	0.512						3.7679
-89.01	0.654						3.7599
-89.11	0.797						3.7519
-87.57	0.937						3.7439
-86.81	0.955						3.7359
-86.87	0.943						3.7279
-87.11	0.913						3.7199
-87.57	0.861	•	•	•	•	•	3.7119
-88.11	0.787						3.7039
-88.77	0.693						3.6959













## 6.0 A SUMMARY OF OTHER AREAS OF ENDEAVOR

### 6.1 The Need For Integration of Antenna Studies With Processor Studies

The University of Texas at Austin Applied Research Laboratory has been developing a computer model of the SMR radar processor with the goal of specifying hardware requirements, data rates, etc., for the SMR system. Communication lines between ARL and PSL have been set up during the latter part of the Phase I contract period in order to integrate PSL's antenna model and ARL's processor model into a more realistic total SMR system simulation model which will eventually be capable of accurately predicting the performance of the overall system. Prior to this interchange of information, both ARL and PSL were using well-developed models of their respective subsystems, but both teams had made somewhat unrealistic assumptions about the quality of the other group's components. Both groups were assuming "perfect", ideal performance characteristics from the other's subsystem.

It is felt that a continuing exchange of ideas, results, and data will yield two important benefits in the shuttle study effort. First, the very act of exchanging data demands some familiarity with the roles which the antenna, radar, and processor play and the way they interact with each other and the rest of the system. Too often this understanding is lacking, resulting in a poorly designed system composed of independently well-designed subsystems. By "closing the loop" with a free exchange of information, it will be possible to provide NASA/JSC with information that will be able to predict accurately the total system performance. The ARL or PSL models standing alone cannot do this.

Secondly, any constraints upon the antenna imposed by the rest of the system can be identified, and vice versa. This will allow bounds to be set on optimum design requirements based on errors and uncertainties within other system components. For example, a user may desire a 25 meter ground cell resolution over a 100 km swath, but because of filtering problems in the range compressor, it may not be feasible to implement so fine a resolution. As a consequence of this radar restriction, it may be possible to relax

beam pointing accuracy requirements. By exploring all such constraints it should be possible to "box in" a feasible region of acceptable operation. By making hypothetical tradeoffs within this region, the most cost-effective system can be specified.

Passing large amounts of data between two widely separated locations can become a formidable task unless computer compatible magnetic tapes or disks are used as the medium of communications. An auxiliary computer program was written to create data tapes from the output of the PSL antenna simulation program. The contents of these tapes is discussed in the next section.

## 6.2 The ARL SMRA Performance Data Tape

The details of the data tape contents, format, etc., are described in the Interface Control Document entitled, "The Shuttle Imaging Radar Antenna Computer Simulation Data Tape" of August 27, 1976. Hence, only a brief description will be given here.

The (one-way) far-field radiation pattern of an array antenna may be written as the product of three terms.

$$\bar{E}(r, \theta, \phi) = E_0(r) f(\theta, \phi) \bar{g}(\theta, \phi) \quad (6-1)$$

$$E_0(r) = \frac{-j\omega\mu e^{-jkr}}{4\pi r} \quad (6-2)$$

where:

$f(\theta, \phi)$  = antenna array factor

$\bar{g}(\theta, \phi)$  = vector element pattern factor

Only data for  $|f\bar{g}|$  has been stored, since the  $E_0(r)$  term may be calculated without regard to  $\theta$  and  $\phi$ . If polarization information is needed, it may be added quite easily.

A coordinate transformation has been performed to simplify calculations. Rather than store  $|f(\theta, \phi) \bar{g}(\theta, \phi)|$ , let

$$\begin{aligned} u &= \sin\theta \cos\phi \\ v &= \sin\theta \sin\phi \end{aligned} \quad (6-3)$$

The advantage of storing  $f$  and  $g$  as functions of  $u$  and  $v$  is that more data points are clustered within the main beam. However, even in the  $(u,v)$ -coordinate system, the narrow azimuth beam width of the SMR antenna demands rather dense discretization. To minimize interpolation errors, at least twenty points on either side of the main beam (in both  $u$  and  $v$ ) are needed. This requirement makes storage of the entire  $(u,v)$ -space or, equivalently,  $(\theta,\phi)$ -space prohibitive. Therefore, only the main beam and first two side lobes have been included. (All other side lobes are nominally 20 dB or more below the main beam maximum.) Data may be recovered quite easily by using the procedure described in the ICD.

### 6.3 SMRA Subarray Preliminary Specifications

The purpose of constructing subarray panels is two-fold:

1. To produce realistic simulation and measurement of reduced-size array behavior and to extend this to a prediction of full-size array behavior.
2. To verify the ability of near-field antenna pattern measurement techniques to measure full-scale SMRA characteristics, particularly in gain, beam coincidence, and cross-polarization levels at X-band.

The specifications on these test panels are based on an estimate of the tests and measurements that will be required to obtain the above results.

Objective (1) may be achieved by measurement of antenna characteristics (such as beam shape, footprint coincidence, etc.) under laboratory-simulated thermal and mechanical stress. Phase I studies have shown this to be critical, particularly at X-band. Once this baseline simulation has been done, systematic errors in the mechanical flatness of the subarray can be induced by mounting the subarray in a standard rigid jig; antenna pattern would then be measured for the subarray under these simulated stress conditions, which would be used to verify the ability of the computer simulation mode (developed under Phases I and II) to predict the actual measured antenna pattern distortions. Finally, from the measured data and the verified computer simulation, it should be possible to predict the performance of the full-scale SMR antenna, under response to space condition thermal stresses, panel unfolding mechanism accuracy, etc.

To check the above mentioned near-field measurements, the near-field antenna patterns (measured at NBS) will be compared with the far-field measurements taken at PSL.

The measurement specifications and their justifications that are listed here are based on data presently available, and as such are only preliminary requirements. As more information regarding antenna behavior is obtained through interactive graphics simulations of various space environmental conditions, these specifications will be refined. Final specifications for the subarrays will be delivered at the close of Phase II.

#### 6.3.1 Measurements Necessary to Obtain Desired Data

Both near-field and far-field pattern measurements should be performed to verify the baseline electrical performance of each subarray. Near-field tests will be conducted by NBS (Boulder), while conventional far-field testing will be done using the 3000' PSL antenna range. A carefully defined set of electrical performance tests will be issued early in Phase III. The purposes of using two independent tests are: 1) to compare conventional far-field testing techniques to the near-field techniques for the subarray and 2) to establish the testing accuracy and capability of the near-field technique, when used for measuring the full-scale SMRA.

Thermal tests will be performed on the subarrays, with measurements of mechanical flatness performed simultaneously. Ideally, the goal of the thermal tests would be to determine the expected thermal response of the full-size antenna plus support structure. However, it has been indicated that the design of a support structure that would replicate the performance of the full-size structure will increase the subarray cost to a prohibitively high level. Consequently, at this time it is felt that it would be more cost-effective to determine the thermal gradient - mechanical deformation relationships between each subarray and a common support structure. After this data has been analyzed, separate specifications can be made for the support structure.



To provide data on the effects of mechanical deformation on antenna pattern parameters, an instrumented jig for the subarrays should be obtained so that simulated random and systematic mechanical errors may be induced into the subarray surface. To obtain pattern data, these tests should be performed simultaneously with the NBS near-field measurements. In this way, the effects of mechanical deformations on antenna performance which were predicted by the computer simulation can be verified, and an extrapolation can later be made to the performance of the full-size SMRA.

The results of these tests will be entered into a more advanced version of the SMRA simulation model as representative of space environmental thermal effects, with predictions made of pattern degradation, beam pointing inaccuracy, etc. Consequently, consideration should be given to the form of measurement data storage. It is our opinion that the most efficient method of managing large amounts of numerical information is to generate computer compatible digital and/or analog magnetic tapes for each test result.

### 6.3.2 Requirements Imposed By Measurement Facilities

In preliminary discussions with Hughes and Ball Brothers, a 6' x 6' area subarray was used as an approximate size. This size estimation was made using the following criteria: 1) the area was electrically large enough to allow meaningful electrical and thermal tests which could with a computer simulation model, be used to predict full-size array performance, and 2) the antenna effective aperture was small enough to conveniently make pattern tests using both near-field and far-field techniques.

The principal concern of (1) is to minimize the edge effects of the smaller subarray so that a better prediction can be made for the full-size SMRA. There are some guidelines in the literature<sup>†</sup>, but even by following these suggestions, the subarrays would still be too large for meaningful far-field measurements. Therefore, criterion (2) is the most restrictive.

<sup>†</sup> c.f. N. Amitay, V. Galindo, and C.P. Wu, Theory and Analysis of Phased Array Antennas, John Wiley and Sons Inc., (New York): 1972, pp. 426-430.

The rule-of-thumb used by many antenna engineers for pattern testing is that the separation between the source and test antenna be greater than  $2D^2/\lambda$ , where  $D$  is the test antenna maximum dimensions. This criterion corresponds to a  $\lambda/16$  path length difference between the source antenna and the extrema of the test antenna. For precise measurement of null depths and side lobe levels, several times this distance may be necessary. The longest leg of the PSL antenna range is 3000', which corresponds to a maximum antenna dimension of 35' at 1.2 GHz and 12.27' at 9.8 GHz. Using four times the rule-of-thumb distance corresponds to 17.5' at 1.2 GHz and 6.135' at 9.8 GHz. This criterion ( $8D^2/\lambda$ ) should provide sufficient accuracy (equivalent to a  $5.625^\circ$  departure from planeness of the incident spherical wave) for all necessary measurements. Therefore, six feet is the upper bound on the maximum dimension (azimuth) of the X-band subarray.

Since the subarrays are to replicate the design approaches of the competing full-size arrays, the azimuth dimensions on the L-band section of the subarray should be 6' also. The elevation dimension of the arrays are open at this time, but for ease of handling, cost minimization, etc., it is proposed that it also be limited to six feet. This makes practical sense, because increasing the elevation dimension will not decrease the edge effects in the azimuth dimensions. Consequently, the most cost-effective design for the measurements which are necessary would be a square subarray.

The 6' x 6' subarray size can be accommodated by the other test facilities needed for the measurements described in Section 6.3.1. The near-field facility at NBS (Boulder, Colorado) can measure both gain and directivity of this size antenna with an uncertainty claim of better than 0.2 dB at the  $3\sigma$  level, at both L- and X-bands. Facilities exist at NASA/JSC and elsewhere that can perform meaningful thermal and mechanical tests on a 6' x 6' structure.

### 6.3.3 A Summary of The Preliminary Specifications

The preliminary specifications for the SMRA test panels, based on the above considerations, are presented in Table 3.

**TABLE 3. Preliminary specifications for the SMRA test panels.**

1. The size shall not exceed 6' x 6'.
2. The weight shall not exceed 200 lbs.
3. The test panel shall operate at 1.5 GHz and 9.0 GHz.
4. Both horizontal and vertical polarization shall be available at each frequency.
5. The cross-polarized component shall not exceed -25 dB with respect to the principally-polarized component.
6. Two modules shall be present at both frequencies to demonstrate beamwidth switching.
7. This VSWR shall not exceed 1.3 in any mode of operation.
8. The maximum side lobe level shall not exceed 12 dB.
9. All array elements, feedlines, and the electrical design approach shall be the same as that which will be used on the full sized antenna.

## APPENDIX

This appendix contains the computer listing of the Shuttle Multispectral Radar Antenna Simulation Program. This listing was obtained by compiling the program on the Physical Science Laboratory/New Mexico State University IBM 370/135 computer using DOS FORTRAN IV.

While the majority of this program can be run on any compatible IBM machine, the user is cautioned to examine all subprograms carefully for calls to functions and subroutines that may not be standard software items. Special attention is called to the subprograms PLOT, AXIS, LINE, and NUMBER.

005 FORTRAN IV 360N-FO-479 3-9 MAINPGM DATE 12/17/76 TIME 09.16.04

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C MAIN PROGRAM FOR SPACE SHUTTLE EARTH RESOURCES IMAGING RADAR ANTENNA

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C DATE: 2 AUGUST 1976  
C REVISED: 23 AUGUST 1976

C VERSION 2.3

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0001 REAL APLLOT(151,151),CON(10),K,MU,XPRINT(501,2),YPRINT(501,2),  
\$XPLOT(501),YPLOT(501),ZAVG(24,12),ALPHAX(24,12),ALPHAY(24,12),  
\$ANGLE(3,3)

0002 REAL KX,KY

0003 REAL PHSX(24,12),AMAG(24,12),APHS(24,12),PHSY(24,12)

0004 COMPLEX AF,AF1,PAT,CEXP,J

0005 INTEGER N(80),DELIM,PR1,PR2,PR3,PL1,PL2,PL3,TP1,TP2

0006 INTEGER DATE(5)

0007 EQUIVALENCE (XPRINT(1,1),YPRINT(1,1)),(XPLOT(1),YPLOT(1))

0008 COMMON /ANG/ YAW,TILT,TWIST,CTW,STW,CT,ST,CYAW,SYAW

0009 COMMON /IO/ PRI,PR2,PL1,PL2,PL3,TP1,TP2,NCON,CON,STARTX,STOPX,  
\$DELTA,STARY,STOPY,DELTA,CONLOW,CONMAX,NPTSX,NPTSY,FLOOR,DASH

0010 COMMON /ARRAY/ APLLOT

0011 COMMON /SYS/ REARTH,RE3,ALT,ALT3,ANGLE

0012 COMMON /ANT/ MSECT,NSECT,MEL,NEL, SX,SY, PX, PY, IHV, K, OMU4PI, ZAVG,  
\$ALPHAX, ALPHAY, KX, KY, PXSECT, PYSECT

0013 COMMON /ELCTRC/ PHSX, PHSY, AMAG, APHS

DDOS FORTRAN IV 360N-FO-479 3-9      MAINPGM      DATE 12/17/76      TIME 09.16.04

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0014      DATA J/(0.,1.)//
0015      DATA PI,MU,RADDEG/3.14159265E0,12.56637E-7,0.01745329E0/
0016      DATA DELIM /'E.'/
0017      FLCOR=-30.0

C
0018      C.....INPUT USER PARAMETERS
0019      1000 READ(1,1,END=9999) NUMPAT,DATE
0020      1 FORMAT(15,1X,5A4)
0021      WRITE(3,2) NUMPAT,DATE
0022      2 FORMAT(1H1,12X,'PATTERN ',13,15X,'SPACE SHUTTLE IMAGING RADAR ANTE
0023      $MNA SIMULATION PROGRAM',14X,5A4////)
0024      9 IYES=0
0025      READ(1,3) NAR
0026      3 FORMAT(80A1)
0027      C.....CHECK FOR DELIMITER
0028      DO 10 I=1,80
0029      IF(NAR(I).EQ.DELIM) GO TO 11
0030      10 CONTINUE
0031      IYES=1
0032      I=81
0033      I=I-1
0034      WRITE(3,4) (NAR(JJ),JJ=1,I)
0035      4 FORMAT(20X,80A1)
0036      IF(IYES.EQ.1) GO TO 9
0037      WRITE(3,5)
0038      5 FORMAT(////)
0039      C.....END OF NARRATIVE
0040      6 FORMAT(8F10.0)
0041      7 FORMAT(2I5)
0042      8 FORMAT(5I1)

C
C      CALL SUBROUTINES FOR FURTHER OUTPUT CALCULATIONS
C
0043      CALL ORBIT(NUMPAT,FREQ)
0044      CALL ANTENA(NUMPAT)
0045      CALL MECH(NUMPAT)
0046      CALL ELEC(MSECT,NSECT,NUMPAT)

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```

0043 CALL OUTPUT(NUMPAT)
C
C.....CALCULATE RUN PARAMETERS
C
      K=2.*PI*FREQ/30.0
      OMEGA=2.*PI*FREQ/IE9
      OMU4PI=OMEGA*MU/(4.*PI)
      DXD=DELTAX/RADDEG
      DYD=DELTAY/RADDEG
C
      WRITE(3,200) NUMPAT
      200 FORMAT(20X,'SUBARRAY DATA SUMMARY FOR PATTERN ',I3,' : /
      $1X,'AREA',8X,'XCENT',9X,'ZAVG',8X,'ALPHAX',7X,
      $'ALPHAY',8X,'AMAG',9X,'APHS',9X,'PHSX',9X,'PHSY')
      IAREA=0
      DO 201 N=1,NSECT
      YCENT=SY*DEL*(N-FLOAT(NSECT+1))*0.5)
      DO 201 M=1,MSECT
      XCENT=SX*DEL*(M-FLOAT(MSECT+1))*0.5)
      IAREA=IAREA+1
      P1=ZAVG(M,N)
      P2=ALPHAX(M,N)*57.295
      P3=ALPHAY(M,N)*57.295
      P4=APAG(M,N)
      P5=APHS(M,N)*57.295
      P6=PHSX(M,N)*57.295
      P7=PHSY(M,N)*57.295
      WRITE(3,202) IAREA,XCENT,YCENT,P1,P2,P3,P4,P5,P6,P7
      202 FORMAT(2X,I3,6X,F8.2,6X,F8.2,6X,F7.2,6X,F7.3,6X,F7.3,6X,F7.4,
      $6X,F7.2,6X,F7.2,6X,F7.2)
      201 CONTINUE
C
C.....CALCULATE BEAM CENTER
C
      KX=K*SX
      KY=K*SY
      CALL GENER
      UP=-PX/KX

```

COS FORTRAN IV 360N-FO-479 3-9 MAINPGM DATE 12/17/76 TIME 09.16.04

```
0071 VP=-PY/XY
0072 IF (UP.EQ.0. .AND. VP.EQ.0.) GO TO 20
0073 PHIP=ATAN2(VP,UP)
0074 GO TO 21
0075 20 PHIP=0.
0076 21 CONTINUE
0077 WP=SQRT(1.0-(UP*UP+VP*VP))
0078 THETAP=APCOS(WP)
C.....UNTWIST
C
0079 PHIP=PHIP+TWIST
0080 XP=SIN(THETAP)*COS(PHIP)
0081 YP=SIN(THETAP)*SIN(PHIP)
0082 ZP=COS(THETAP)
C....."UNTILT"
C
0083 X=XP
0084 Y=CT*YP+ST*ZP
0085 Z=-ST*YP+CT*ZP
C....."UNYAW"
C
0086 XP=CYAW*X-SYAW*Y
0087 YP=SYAW*X+CYAW*Y
0088 ZP=Z
C.....CALCULATE ANGLES
C
0089 RP=SQRT(XP*XP+YP*YP+ZP*ZP)
0090 THETAP=ARCCOS(ZP/RP)
0091 IF (XF.EQ.0. .AND. YP.EQ.0.) GO TO 22
0092 PHIP=ATAN2(YP,XP)
0093 GO TO 23
0094 22 PHIP=0.0
0095 23 CONTINUE
C
```





DCS FORTRAN IV 360N-FC-479 3-9 MAINPGM DATE 12/17/76 TIME 09.16.04

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0121 IF(CABS(AF).LT.1E-10) AF=(1E-10,0.)
0122 APLOT(MM,NN)=20.*ALOG10(CABS(AF))-BIG
0123 40 CONTINUE
C
0124 IF(PL2.EQ.1) CALL PLOT2(NPTSX,NPTS,CON,NCON,NUMPAT,-1000.)
0125 IF(PR2.EQ.1) CALL PATCON(APLOT,CONLOW,CONMAX,STARTX/RADDEG,
$STOPX/RADDEG,STARTY/RADDEG,STOPY/RADDEG,NUMPAT)
0126 IF(PL3.NE.1) GO TO 50
0127 DO 45 M=1,NPTSX
0128 DO 45 N=1,NPTS
0129 IF(APLOT(M,N).LT.FLOOR) APLOT(M,N)=FLOOR
0130 45 CONTINUE
0131 CALL PLOT3(NPTSX,NPTS,NUMPAT)
C
C.....ONE DIMENSIONAL PLOTS
C
0132 50 CONTINUE
0133 IF(PRI+PL1.EQ.0) GO TO 51
C
C CALCULATE PROFILE ALONG X-AXIS AT Y=YC
C
0134 DX=(STOPX-STARTX)/500.0
0135 YR=FLAT
0136 XR=STARTX-DX
0137 DO 60 MM=1,501
0138 XR=XR+DX
0139 AF=PAT(XR,YR)
0140 XPRINT(MM,1)=XR/RADDEG
0141 XPRINT(MM,2)=CABS(AF)/BIG1
0142 IF(XPRINT(MM,2).LT.1E-10) XPRINT(MM,2)=1E-10
0143 XPLAT(MM)=20.*ALOG10(XPRINT(MM,2))
0144 IF(PRI.NE.1) GO TO 63
0145 WRITE(3,114) FLAT0
0146 114 FORMAT(1H1,25X,'X-AXIS PROFILE PLOT ALONG ',F8.3,' DEGREES LATITUD
$E.')
C
0147 CALL PROFIL(XPRINT,501,NUMPAT)
0148 63 CONTINUE
0149 IF(PL1.NE.1) GO TO 64

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DCS FORTRAN IV 360N-FC-479 3-9 MAINPGM DATE 12/17/76 TIME 09.16.04

0150 CALL PLOT1(STARTX/RADDEG,STOPX/RADDEG,500,1,FLATO,NUMPAT,XPLOT)  
0151 64 CONTINUE

C  
C.....CALCULATE PROFILE ALONG Y-AXIS AT X=FLAT  
DY=(STOPY-STARTY)/500.0  
XR=FLUNG  
YR=STARTY-DY  
DO 70 MM=1,501  
YR=YR+DY  
AF=PIAT(XR,YR)  
YPRINT(MM,1)=YR/RADDEG  
YPRINT(MM,2)=CABS(AF)/BIG1  
IF(YPRINT(MM,2).LT.1E-10) YPRINT(MM,2)=1E-10  
70 YPLOT(MM)=20.\*ALOG10(YPRINT(MM,2))

C  
C.....OUTPUT SEQUENCE  
IF(PI.NE.1) GO TO 80  
WRITE(3,115) FLONGD  
115 FORMAT(1H1,25X,'Y-AXIS PROFILE PLOT ALONG',F8.3,'LONGITUDE.')

C.....PLOTTER PLOTS  
80 CONTINUE  
IF(PI.NE.1) GO TO 90  
CALL PLOT1(STARTY/RADDEG,STOPY/RADDEG,500,2,FLONGD,NUMPAT,YPLOT)  
90 CONTINUE  
51 CONTINUE  
95 CONTINUE  
C.....NEW SEARCH CODE  
CALL PLOT(0.,0.,-3)  
GO TO 1000

C  
C  
C 9999 CONTINUE  
WRITE(3,117)  
117 FORMAT(20X,'NORMAL TERMINATION.')

0174  
0175  
0176  
0177  
0178

DCS FORTRA IV 360N-FO-479 3-9 ORBIT DATE 12/17/76 TIME 09.16.34

```

0001 SUBROUTINE ORBIT(NUMPAT,FREQ)
0002 DATA RADDEG /0.01745329E0/
0003 REAL ANGLE(3,3)
0004 COMMON /ANG/ YAW,TILT,TWIST,CTW,STW,CT,ST,CYAW,SYAW
0005 COMMON /SYS/ REARTH,RE3,ALT,ALT3,ANGLE
0006 REARTH = 6370.0
0007 RE3=REARTH*IE3
0008 READ(1,6) FREQ
0009 READ(1,6) ALT
0010 6 FORMAT(8F10.0)
0011 ALT3=ALT*IE3
0012 PEAD(1,6) YAWD,TILTD,TWISTD
0013 TWIST=RADDEG*TWISTD
0014 TILT=RADDEG*TILTD
0015 YAW=RADDEG*YAWD
0016 CTW=COS(TWIST)
0017 STW=SIN(TWIST)
0018 CT=COS(TILT)
0019 ST=SIN(TILT)
0020 CYAW=COS(YAW)
0021 SYAW=SIN(YAW)

```

C

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0022 ANGLE(1,1)=CTW*CYAW-SYAW*CT*STW
0023 ANGLE(1,2)=SYAW*CTW+CYAW*CT*STW
0024 ANGLE(1,3)=-ST*STW
0025 ANGLE(2,1)=-STW*CYAW-SYAW*CT*CTW
0026 ANGLE(2,2)=CYAW*CT*CTW-SYAW*STW
0027 ANGLE(2,3)=-ST*CTW
0028 ANGLE(3,1)=-SYAW*ST
0029 ANGLE(3,2)=CYAW*ST
0030 ANGLE(3,3)=CT

```

C

C.....OUTPUT SYSTEM INFORMATION

C

```

0031 WRITE(3,100) FREQ
0032 100 FORMAT(20X,'SYSTEM INFORMATION: '/25X,'FREQUENCY = ',F6.3,' GHZ. ')
0033 WRITE(3,101) YAWD,TILTD,TWISTD,ALT
0034 101 FORMAT(25X,'YAW = ',F8.3,' DEGREES'/25X,'TILT = ',F8.3,' DEGREES'/'

```

DOS FORTRAN IV 360N-FC-479 3-9 ORBIT DATE 12/17/76 TIME 09.16.34

\$25X, 'TWIST = ', F8.3, ' DEGREES', /25X, 'ALTITUDE = ', F8.3, ' KM.' '////)

C

0035  
C036

RETURN  
END

```

0001 SUBROUTINE ANTENA(NUMPAT)
C
C THIS ROUTINE INPUTS AND CALCULATES APPROPRIATE ANTENNA PARAMETERS
C FOR THE SPACE SHUTTLE SYNTHETIC IMAGING RADAR ANTENNA.
C
C FINALLY, ALL ANTENNA PARAMETERS ARE OUTPUTTED.
C
C-----ANTENNA CONFIGURATION -- TWO-DIMENSIONAL ARRAY-----C
C
REAL ZAVG(24,12),ALPHAX(24,12),ALPHAY(24,12),K
REAL KX,KY
COMMON /ANT/ MSECT,NSECT,MEL,NEL,SX,SY,PX,PY,IHV,K,OMU4PI,ZAVG,
$ALPHAX,ALPHAY,KX,KY,PXSECT,PYSECT
C
C
READ(1,1) MSECT,NSECT
READ(1,1) NX,NY
1 FORMAT(8I5)
MEL=NX/MSECT
NEL=NY/NSECT
READ(1,2) SX,SY
2 FORMAT(8F10.0)
READ(1,2) PXD,PYD
PX=PXD*0.01745329EO
PY=PYD*0.01745329EO
READ(1,1) IHV
WRITE(3,3) NUMPAT
3 FORMAT(20X,'ANTENNA PARAMETERS FOR SIMULATION NUMBER ',I3)
IF(IHV.EQ.0) WRITE(3,100)
IF(IHV.EQ.1) WRITE(3,101)
IF(IHV.EQ.2) WRITE(3,102)
100 FORMAT(25X,'ELEMENT TYPE: ISOTROPIC')
101 FORMAT(25X,'ELEMENT TYPE: HORIZONTAL DIPOLE')
102 FORMAT(25X,'ELEMENT TYPE: VERTICAL DIPOLE')
WRITE(3,4) NX,NY
4 FORMAT(25X,'NUMBER OF ELEMENTS (X,Y) = (',I3,', ',I3,',)')
WRITE(3,5) SX,SY

```

```
CGS FORTRAN IV 360N-FO-479 3-9      ANTENA      DATE 12/17/76      TIME 09.16.45
0027      5 FORMAT(25X,'INTERELEMEN SPACING (CM.): ',F7.4,' ',F7.4)
0028      WRITE(3,6) PXD,PYD
0029      6 FORMAT(25X,'INTERELEMEN PHASE SHIFT (DEG.): ',F7.4,' ',F7.4//)
0030      RETURN
0031      END
```

DDS FORTRAN IV 360N-FC-479 3-9 MECH DATE 12/17/76 TIME 09.16.56

```

0001 SUBROUTINE MECH(NUMPAT)
0002 REAL ZAVG(24,12),ALPHAX(24,12),ALPHAY(24,12),K
0003 REAL A(3,3)
0004 REAL KX,KY
0005 REAL WARP(201)
0006 COMMON /SYS/ RE,REB,ALT,ALT3,A
0007 COMMON /ANT/ MSECT,NSECT,MEL,NEL,SX,SY,PX,PY,IHV,K,OMU4PI,ZAVG,
    $ALPHAX,ALPHAY,KX,KY,PXSECT,PYSECT

```

C  
C

```

0008 CALL MISC(MSECT,NSECT,WARP)
0009 WRITE(3,100) NUMPAT
0010 100 FORMAT(20X,'DEFORMATION DATA FOR SIMULATION ',I3,' : //')
0011 MSFCT1=NSECT+1
0012 MSFCTI=MSECT+1
0013 DO 50 II=1,NSECT1
0014 I=NSECT1-II+1
0015 II=(I-1)*MSECT1+1
0016 III=I*MSECT1
0017 WRITE(3,101) (WARP(IJ), IJ=II,III)
0018 101 FORMAT(10(3X,F7.2))
0019 WRITE(3,105)
0020 105 FORMAT(/)
0021 50 CONTINUE
0022 WRITE(3,104)

```

C  
C

```

0023 PXSECT=MEL*PX
0024 PYSECT=NEL*PY
0025 XLEN=SX*MEL
0026 YLEN=SY*NEL
0027 DX2=0.5/XLEN
0028 DY2=0.5/YLEN

```

C  
C

```

0029 104 FORMAT(////)
0030 TAREA=0
0031 DO 30 N=1,NSECT

```

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR



DDS FORTRAN IV 360N-FO-479 3-9 MECH DATE 12/17/76 TIME 09.16.56

```
0032 YCENT=YLEN*(N-FLOAT(NSECT+1))*0.5)
0033 DO 30 M=1,MSECT
0034 XCENT=XLEN*(M-FLOAT(MSECT+1))*0.5)
0035 IAREA=IAREA+1
0036 NUM=(IAREA+1)*N
0037 AO=WARP(NUM)
0038 A1=WARP(NUM+1)
0039 A2=WARP(NUM+1+NSECT)
0040 A3=WARP(NUM+2+MSECT)
C
0041 B0=(A0+A1+A2+A3)*0.25
0042 B1=DX2*(-A0+A1-A2+A3)
0043 B2=DY2*(-A0-A1+A2+A3)
0044 B3=DX2*DY2*4*(A0-A1-A2+A3)
C
0045 ZAVG(M,N)=B0
0046 ALPHA(M,N)=ATAN(B1)
0047 ALPHAY(M,N)=ATAN(B2)
0048 30 CONTINUE
0049 WRITE(3,104)
0050 RETURN
0051 END
```

005 FORTRAN 1. 360N-FO-479 3-9 MISC DATE 12/17/76 TIME 09.17.09

```
0001 SUBROUTINE MISC(MSECT,NSECT,WARP)
0002 REAL WARP(201)
0003 NWARP=(MSECT+1)*(NSECT+1)
0004 READ(1,2) (WARP(I),I=1,NWARP)
0005 2 FORMAT(8F10.0)
0006 RETURN
0007 END
```

005 FORTRAN IV 360N-FO-479 3-9 ELEC DATE 12/17/76 TIME 09.17.17

```

0001 SUBROUTINE ELEC(MSECT,NSECT,NUMPAT)
0002 REAL PHSX(24,12),AMAG(24,12),APHS(24,12),PHSY(24,12)
0003 COMMON /ELCTRC/ PHSX,PHSY,AMAG,APHS
0004 MNSECT=(MSECT+1)*(NSECT+1)
0005 DO 10 J=1,NSECT
0006 DO 10 I=1,MSECT
0007 10 READ(1,11) PHSX(I,J),PHSY(I,J),AMAG(I,J),APHS(I,J)
0008 11 FORMAT(3F10.0)
C
C
0009 WRITE(3,12) NUMPAT
0010 12 FORMAT(20X,'ELECTRICAL DATA FOR SIMULATION ',I3,' :')
0011 J=NSECT+1
0012 DO 20 JJ=1,NSECT
0013 J=J-1
0014 WRITE(3,25) (I,J,I=1,MSECT)
0015 WRITE(3,21) (PHSX(I,J),I=1,MSECT)
0016 WRITE(3,22) (PHSY(I,J),I=1,MSECT)
0017 WRITE(3,23) (AMAG(I,J),I=1,MSECT)
0018 WRITE(3,24) (APHS(I,J),I=1,MSECT)
0019 WRITE(3,104)
0020 20 CONTINUE
0021 21 FORMAT(5X,'PHSX:',10(3X,F7.2))
0022 22 FORMAT(5X,'PHSY:',10(3X,F7.2))
0023 23 FORMAT(5X,'AMAG:',10(3X,F7.2))
0024 24 FORMAT(5X,'APHS:',10(3X,F7.2))
0025 25 FORMAT(10X,10(3X,'(',I2,',',I2,')'))
0026 DO 30 J=1,NSECT
0027 DO 30 I=1,MSECT
0028 PHSX(I,J)=PHSX(I,J)*0.017453293E0
0029 PHSY(I,J)=PHSY(I,J)*0.017453293E0
0030 APHS(I,J)=APHS(I,J)*0.017453293E0
0031 30 CONTINUE
0032 WRITE(3,104)
0033 104 FORMAT(////)
0034 RETURN
0035 END

```

0001 FUNCTION PAT(XR,YR)

C PAT COMPUTES THE COMPLEX-VALUED PATTERN FROM A PIECEWISE  
 C BILINEAR RECTANGULAR ARRAY.

C ANTENNA PARAMETERS ARE PASSED BY COMMON BLOCK ANT.  
 C OTHER SYSTEM PARAMETERS ARE PASSED BY COMMON BLOCK SYS.

C WRITTEN BY: E. L. COFFEY  
 C DATE: 26 JULY 1976

0002 COMPLEX PAT,CEXP,J  
 0003 COMPLEX AF2  
 0004 REAL ZAVG(24,12),ALPHAX(24,12),ALPHAY(24,12),K  
 0005 REAL KX,KY  
 0006 REAL PHSX(24,12),AMAG(24,12),APHS(24,12),PHSY(24,12)  
 0007 REAL A(3,3)

0008 COMMON /SYS/ RE,RE3,ALT,ALT3,A  
 0009 COMMON /ANT/ MSECT,MSECT,MEL,NEL,SX,SY,PX,PY,IHV,K,OMU4PI,  
 \$ZAVG,ALPHAX,ALPHAY,KX,KY,PXSECT,PYSECT  
 0010 COMMON /ELCTRC/ PHSX,PHSY,AMAG,APHS

0011 DATA J/(0.,1.)/

C.....TRANSLATION FROM EARTH SURFACE TO UNROTATED ANTENNA.

0012 CYR=COS(YR)  
 0013 XP=-RE3\*SIN(XR)\*CYR  
 0014 YP=RE3\*SIN(YR)  
 0015 ZP=RE3\*(1.-COS(XR)\*CYR)+ALT3

C.....ANTENNA YAW,TILT,TWIST  
 0016 XPP=A(1,1)\*XP+A(1,2)\*YP+A(1,3)\*ZP  
 0017 YPP=A(2,1)\*XP+A(2,2)\*YP+A(2,3)\*ZP  
 0018 ZPP=A(3,1)\*XP+A(3,2)\*YP+A(3,3)\*ZP

```

0019 RPP=SQRT(XPP*XPP+YPP*YPP+ZPP*ZPP)
0020 U=XPP/RPP
0021 V=YPP/RPP
0022 ALPHA1=ARCOS(U)
0023 ALPHA2=ARCOS(V)
0024 AFI=CMU4PI/RPP

```

C  
C..... TRANSLATE TO CENTER OF SUBARRAY  
C

```

0025 DX=SX*MEL
0026 DY=SY*NEL
0027 AF2={0.,0.}
0028 YCENT=-FLOAT(NSECT+1)/2.0*DY
0029 DO 10 N=1,NSECT
0030 YCENT=YCENT+DY
0031 XCENT=-FLOAT(MSECT+1)/2.0*DX
0032 DO 10 M=1,MSECT
0033 XCENT=XCENT+DX
0034 UP=COS(ALPHA1-ALPHA2-ALPHAX(M,N))+PHSX(M,N)/KX
0035 VP=COS(ALPHA2-ALPHAY(M,N))+PHSY(M,N)/KY
0036 WP=SQRT(1.-((UP*UP+VP*VP)))
0037 PHASE=K*(XCENT*UP+YCENT*VP+ZAVG(M,N)*WP)+M*PXSECT+N*PYSECT
    $+APHS(M,N)
0038 AF2=AF2+AF1*CMPLX(COS(PHASE),SIN(PHASE))*AF(UP,VP)
    $*AMAG(M,N)

```

10 CONTINUE

```

0039 C
0040 C
0041 PAT=AF2
0042 RETURN
0043 END

```

```

0001 SUBROUTINE GENER
0002 REAL ZAVG(24,12),ALPHAX(24,12),ALPHAY(24,12),K
0003 REAL KX,KY
0004 REAL UARRAY(1001),VARRAY(1001)
0005 COMMON /PATRN/ UARRAY,VARRAY,UMAX,VMAX,DU,DV
0006 COMMON /ANT/ MSECT,NSECT,MEL,NEL,SX,SY,PX,PY,IHV,K,DMU4PI,
      $ZAVG,ALPHAX,ALPHAY,KX,KY,PXSECT,PYSECT
C
C.....THIS ROUTINE GENERATES PATTERN DATA FOR THE ARRAYS UARRAY AND VARRAY.
C
      NULNUM=3
      UMAX=6.28318531E0/K/((MEL*SX)*NULNUM
      VMAX=6.28318531E0/K/((MEL*SY)*NULNUM
      IF(UMAX.GT.1.0) UMAX=1.0
      IF(VMAX.GT.1.0) VMAX=1.0
C
      DU=UMAX*1E-3
      DV=VMAX*1E-3
C
C
      DO 10 I=1,1001
      U=(I-1)*DU
      PSIX=KX*U+PX
      PSIX2=0.5*PSIX
      UI=1.0
      IF(PSIX.NE.0.) UI=SIN(MEL*PSIX2)/(MEL*SIN(PSIX2))
      IF(IHV.EQ.1.AND.ABS(U).NE.1.0)UI=UI*COS(1.570796E0*U)/SQRT(1.-U*U)
      UARRAY(I)=UI
C
      V=(I-1)*DV
      PSIY=KY*V+PY
      PSIY2=0.5*PSIY
      VI=1.0
      IF(PSIY.NE.0.) VI=SIN(NEL*PSIY2)/(NEL*SIN(PSIY2))
      IF(IHV.EQ.2.AND.ABS(V).NE.1.0)VI=VI*COS(1.570796E0*V)/SQRT(1.-V*V)
      VARRAY(I)=VI
      10 CONTINUE
C

```

DCS FORTRAN IV 360N-FO-479 3-9      GENER      DATE 12/17/76      TIME 09.17.43

C  
0030      RETURN  
0031      END

REPRODUCIBILITY OF THE  
ORIGINAL PAGE IS POOR

```

0001      FUNCTION AF(U1,V1)
C
C.....THIS FUNCTION CALCULATES THE ARRAY FACTOR OF A SUBPANEL BY
C      "TABLE LOOK-UP" IF U < UMAX AND V < VMAX. OTHERWISE, THE FUNCTION
C      IS COMPUTED IN THE USUAL WAY.
C
C      LINEAR INTERPOLATION IS USED IN THE TABLE LOOK-UP.
C
0002      REAL ZAVG(24,12),ALPHAX(24,12),ALPHAY(24,12),X
0003      REAL KX,KY
0004      REAL UARRAY(1001),VARRAY(1001)
C
0005      COMMON /PATTRN/ UARRAY,VARRAY,UMAX,VMAX,DJ,DV
0006      COMMON /ANT/ MSECT,NSFCT,MEL,NEL,SX,SY,PX,PY,IHV,K,OMJ4PI,
          $ZAVG,ALPHAX,ALPHAY,KX,KY,PXSECT,PYSECT
C
0007      U=ABS(U1)
0008      V=ABS(V1)
C
0009      IF(U.GT.UMAX) GO TO 10
C.....U IS IN RANGE
0010      FU=U/DU+1.0
0011      IFU=IFIX(FU)
0012      DIFF=FU-IFU
0013      UPAT=(1.0-DIFF)*UARRAY(IFU)+DIFF*UARRAY(IFU+1)
0014      GO TO 15
0015      10 CONTINUE
0016      PSIX2=(KX*U+PX)*0.5
0017      UPAT=SIN(MEL*PSIX2)/(MEL*SIN(PSIX2))
0018      IF(IHV.EQ.1) UPAT=UPAT*COS(1.570796E0*U)/SQRT(1.-U*U)
0019      15 CONTINUE
0020      IF(V.GT.VMAX) GO TO 20
C.....V IS IN RANGE
0021      FV=V/DV+1.0
0022      IFV=IFIX(FV)
0023      DIFF=FV-IFV
0024      VPAT=(1.0-DIFF)*VARRAY(IFV)+DIFF*VARRAY(IFV+1)

```



005 FORTRAN IV 360N-FD-479 3-9 AF DATE 12/17/76 TIME 09.17.55

```
0025 GO TO 25
0026 20 CONTINUE
0027    PSY2=(KY*V+PY)*0.5
0028    VPAT=SIN(NEL*PSY2)/(NEL*SIN(PSY2))
0029    IF(IHV.EQ.2) VPAT=VPAT*COS(1.570796E0*V)/SQRT(1.-V*V)
0030 25 CONTINUE
0031    AF=UPAT*VPAT/(MSECT*NSECT)
0032    RETURN
0033    END
```



```

0029
0030
0031
0032
0033
0034
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0038
0039
0040
0041
0042
0043
0044
0045
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0048
0049
0050
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0052
0053
0054
0055

C
C-----OUTPUT PRINT/PLOT INFORMATION
C
      WRITE(3,103)
003  FORMAT(//'/20X,'PRINT/PLOT INFORMATION: '/25X,'REQUESTED OUTPUT:')
      IF(PR1.EQ.1) WRITE(3,104)
      IF(PR2.EQ.1) WRITE(3,105)
      IF(PL1.EQ.1) WRITE(3,106)
      IF(PL2.EQ.1) WRITE(3,107)
      IF(PL3.EQ.1) WRITE(3,108)
004  FORMAT(30X,'PRINTER PROFILE')
005  FORMAT(30X,'PRINTER CONTOUR')
006  FORMAT(30X,'PLOTTER PROFILE')
007  FORMAT(30X,'PLOTTER CONTOUR')
008  FORMAT(30X,'PLOTTER THREE-C')

C
      IF(PL2.NE.1) GO TO 24
      WRITE(3,109) (CON(I),I=1,NCON)
009  FOPMAT(//'/25X,'CONTOURS TO BE PLOTTED: ',5(F6.2,3X)/49X,5(F6.2,3X))
24  CONTINUE
      IF(PR2+PL2+PL3.GE.1) WRITE(3,110) NPTX,NPTS
      WRITE(3,111) STRXD,STFXD,DXD
011  FORMAT(25X,'STARTX = ',F8.3/25X,'STOPX = ',F8.3/25X,'DELTA X = ',
      $F8.3/)
      WRITE(3,112) STRYD,STPYD,DYD
012  FORMAT(25X,'STARTY = ',F8.3/25X,'STOPY = ',F8.3/25X,'DELTA Y = ',
      $F8.3/)
      6  FORMAT(6F10.0)
      7  FORMAT(2I5)
      8  FORMAT(5I1)
      RETURN
      END

```

ESS FORTRAN IV 360N-FQ-479 3-9      PROFIL      DATE 12/17/76      TIME 09.18.20

```

0001                    SUBROUTINE PROFIL(DATA,NPT,NUMPAT)
C
C                    DATA = ,DATA INPUT (DESTROYED)
C                    NPT = NUMBER OF POINTS
C                    NUMPAT = PATTERN NUMBER
C
0002                    INTEGER SF
0003                    INTEGER OUTPUT(81)
0004                    INTEGER BLANK, PLUS, SLASH, STAR
0005                    REAL DATA(501,2), BOUND(81)
0006                    DATA BLANK, PLUS, SLASH, STAR / ' , + , . , | , * , /

C                    FIND THE RANGE OF DEPENDENT DATA AND SCALE IF NECESSARY
C
C                    IF(NPT.GT.501) GO TO 999
0007                    BIG=-1.E10
0008                    SMALL = 1.E10
0009                    DO 1 J=1,NPT
0010                    IF(DATA(J,2).LT.-60.0) DATA(J,2)=-60.0
0011                    IF(DATA(J,2).LT.SMALL) SMALL=DATA(J,2)
0012                    IF(DATA(J,2).GT.BIG) BIG=DATA(J,2)
0013                    : CONTINUE
0014                    DIFF=ABS(BIG-SMALL)
0015                    SF = 0
0016                    IF(DIFF.LT.1.) GO TO 10
0017                    IF(DIFF.LT.100.)GO TO 21
0018                    DO 2 J=1,10
0019                    IF(DIFF#10.**(-J).GT.100.) GO TO 2
0020                    SF=J
0021                    GO TO 20
0022                    2 CONTINUE
0023                    400 WRITE(3,100)
0024                    100 FORMAT('O YOUR DATA IS TOO LARGE FOR THIS PROGRAM.')
0025                    RETURN
0026                    10 DO 3 J=1,10
0027                    K=11-J
0028                    IF(DIFF#10**K.GT.100.) GO TO 3
0029                    SF=-K
0030                    :

```

DDS FORTRAN IV 360N-FD-479 3-9      PROFIL      DATE 12/17/76      TIME 09.18.20

```

0031      GO TO 20
0032      3 CONTINUE
0033      GO TO 400
0034      20 DO 4 J=1,NPT
0035      4 DATA(J,2) = DATA(J,2)*10.**{(-SF)}
      C
      C      CALCULATE BOUNDS
      C
0036      21 SCALE=DIFF/80.
0037      DO 5 J=1,81
0038      K=J-1
0039      5 BOUND(J)=(BIG-K*SCALE)*10.**{(-SF)}
0040      SFDB=20.*SF
      C
      C      PRINT TITLE
      C
0041      WRITE(3,640) NUMPAT
0042      640 FORMAT(26X,'PATTERN NUMBER ',I5//)
0043      IF (SF.EQ.0) GO TO 200
0044      WRITE(3,4004) SF
0045      4004 FORMAT(53X,'SCALE FACTOR IS 10**',I2//)
0046      200 WRITE(3,650) (BOUND(J),J=1,81,16)
0047      650 FORMAT(3X,'DR.',5X,'REAL',2X,6(F7.4,9X))
0048      DO 6 JI=1,NPT
0049      J=NPT+1-JI
0050      DO 50 K=1,81
0051      50 OUTPUT(K)=BLANK
0052      IF((J-1)/8+8-(J-1)) 62,61,62
0053      61 DO 40 K=1,81,8
0054      40 OUTPUT(K)=PLUS
0055      GO TO 87
0056      62 OUTPUT(1)=SLASH
0057      OUTPUT(81)=SLASH
0058      87 DO 7 K=1,80
0059      IF(DATA(J,2)-GT.BOUND(K)) GO TO 7
0060      IF(DATA(J,2)-LE.BOUND(K+1)) GO TO 7
0061      OUTPUT(K)=STAR
0062      GO TO 69

```



F 05 FORTRAN IV 360N-FO-479 3-9 PATCON DATE 12/17/76 TIME 09.18.35

```
0001 SUBROUTINE PATCON(A , CONLOW, CONMAX, STARTX, STOPX, STARTY, STOPY,
      $NMPAT)
      C
      C THIS ROUTINE CALLS SUBROUTINE CONTUR THREE TIMES TO GENERATE A
      C COMPLETE 151 X 151 TWO-DIM. PLOT.
      C
      C DIMENSION A(151,151)
      C DY=(STOPY-STARTY)/3.0
      C CALL CCNTUR(A ,1, CONLOW, CONMAX, STARTX, STOPX, STARTY, STARTY+DY,
      $NMPAT)
      C CALL CCNTUR(A ,2, CONLOW, CONMAX, STARTX, STOPX, STARTY+DY, STOPY-DY,
      $NMPAT)
      C CALL CCNTUR(A ,3, CONLOW, CONMAX, STARTX, STOPX, STOPY-DY, NMPAT)
      C RETURN
      C END
0002
0003
0004
0005
0006
0007
0008
```

```

0001        SUPROUTINE CONTUR(A,ICODE,CONLOW,CONMAX,STARTX,STOPX,STARTY,STOPY,
          $NUMPAT)
C
C        THIS ROUTINE PRINTS A 51 X 151 CONTOUR PLOT OF THE FOOTPRINT.
C        TO OBTAIN A COMPLETE 151 X 151 PLOT, THIS ROUTINE IS CALLED THREE
C        TIMES BY SUBROUTINE PATCON. THEN THE THREE SEPARATE PLOTS ARE
C        JOINED TOGETHER TO MAKE THE COMPOSITE.
C
C

```

```

0002        REAL A(151,151),LOW(12),HIGH(12),AXIS(11)
0003        INTEGER OUTPUT(101),LEVEL(12),BLANK,DOT

```

```

0004        DATA LEVEL/'0','1','2','3','4','5','6','7','8','9',' ','+'/'
0005        DATA BLANK /' '/

```

```

0006        WRITE(3,10) NUMPAT
0007        10 FORMAT(1H1,37X,'PRINTER CONTOUR PLOT FOR SIMULATION NUMBER ',I3//
          $///)

```

```

C
0008        DELTAX=(STOPX-STARTX)/150.0
0009        DELTAY=(STOPY-STARTY)/10.0
0010        NUMCON=10
0011        CONINT=(CONMAX-CONLOW)/FLOAT(NUMCON-1)
0012        DELCON=CONINT*0.5
0013        DO 71 J=1,NUMCON
0014        LOW(J)=CONLOW+(J-1)*CONINT-DELCON
0015        HIGH(J)=LOW(J)+CONINT+1E-5
0016        71 CONTINUE
0017        LOW(11)=-1E50
0018        HIGH(12)=1E50
0019        HIGH(11)=LOW(1)
0020        LOW(12)=HIGH(NUMCON)

```

```

C
0021        DO 40 I=1,11
0022        40 AXIS(I)=STARTY+(I-1)*DELTAY
0023        WRITE(3,42) (AXIS(I),I=1,11)
0024        42 FORMAT(6X,11(F8.3,2X)/10X,10('|-----|.....'),'(|)')

```





POS FORTRAN IV 360N-FO-479 3-9                    CONTUR                    DATE 12/17/76                    TIME 09.18.44

```
0057                    45 WRITE(3,46) (LEVEL(J),LOW(J),HIGH(J),J=1,12,4)  
0058                    46 FORMAT(2X,3(A1,' ',E14.7,' TO ',E14.7,3X))  
0059                    80 CONTINUE  
0060                    RETURN  
0061                    END
```

0001 SUPROUTINE PLOT1(PSTRT, PEND, IP, CODE, CONST, NJMPAT, PTS)

0002 PROFILE PLOT ROUTINE

0003 WRITTEN BY: E. L. COFFEY  
 0004 DATE: 22 JUNE 1976

0005 INPUT:  
 0006 PSTRT= BEGINNING OF PLOT  
 0007 PEND= END OF PLOT  
 0008 IP= NUMBER OF POINTS TO BE PLOTTED  
 0009 CODE= LABELLING CODE -- 1 FOR X-AXIS, 2 FOR Y-AXIS  
 0010 CONST= "OTHER" AXIS CONSTANT  
 0011 NJMPAT= PATTERN NUMBER...USER DEFINED

0012 INTEGER CODE  
 0013 DIMENSION PTS(501)  
 0014 CALL PLOT(8, 2, 23)  
 0015 CALL FACTOR(0.7)  
 0016 ENIM=NJMPAT

0017 OPEN AXES AND LABEL THEM  
 0018 Y=0.05  
 0019 DO 11 J=1,11  
 0020 X=-5.0+(J-1)\*1.0  
 0021 CALL PLOT(X, -Y, 3)  
 0022 11 CALL PLOT(X, Y, 2)  
 0023 CALL PLOT(5, 0, 3)  
 0024 CALL PLOT(-5, 0, 2)  
 0025 CALL PLOT(0, 5, 3)  
 0026 CALL PLOT(0, 0, 2)  
 0027 X=0.05  
 0028 DO 10 J=1,6  
 0029 Y=0.5+(J-1)\*1.0

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR.

005 FORTAN IV 3604-EN-479 3-9 PLOT1 DATE 03/02/77 TIME 10.00.47

```

0019      CALL PLOT(-X,Y,3)
0020      IO CALL PLOT(X,Y,2)
C
0021      CALL NUMBER(-5.25,-.75,0.125,PSTRT,0.,6)
0022      PMID={PSTRT+PEND)*0.5
0023      CALL NUMBER(-.25,-.75,0.125,PMID,0.,6)
0024      CALL NUMBER(4.75,-.75,0.125,PEND,0.,6)
0025      CALL SYMBOL(-5.0,-1.5,0.125,7HPATYFPN,0.,7)
0026      CALL NUMBER(-3.5,-1.5,0.125,FNUM,0.,-1)
C
0027      IF(CODE.EQ.1) CALL SYMBOL(-2.0,-1.5,0.125,32HX-AXIS PROFILE ALONG
$LATITUDE = ,0.,32)
0028      IF(CODE.EQ.2) CALL SYMBOL(-2.0,-1.5,0.125,33HY-AXIS PROFILE ALONG
$LONGITUDE = ,0.,33)
0029      CALL NUMBER(2.0,-1.5,0.125,CONST,3)
0030      CALL ZIP(1,3,18)
C
C      PLOT IT
C
0031      IF(PTS(1).LT.-50.) PTS(1)=-50.0
0032      FS=PTS(1)/10.+5.5
0033      CALL PLOT(-5.0,FS,3)
0034      PFLTA=10./FLCAT(IP-1)
0035      DO 1 IWI=1,IP
0036      X=-5.0+(IWI-1)*DELTA
0037      Y PTS(IWI)/10.+5.5
0038      IF(Y.LT. 0.5) GO TO 1
0039      CALL PLOT(X,Y,2)
0040      1 CONTINUE
0041      CALL FACTOR(1.0)
0042      CALL PLOT(8.0,-2.0,23)
0043      CALL ZIP(3,3,18)
0044      RETURN
0045      END

```

```

0001      COS FORTRAN IV 360N-FC-479 3-9          PRINT          DATE 03,02/77          TIME 10.01.01
0002      SURROUTINE PLOT2(N,M,CONTOUR,NCN,NUMPAT,NASH)
0003      A= N BY M MATRIX OF DATA POINTS
0004      CONTOUR= ARRAY OF CONTOURS TO BE PLOTTED
0005      NCN= NUMBER OF CONTOURS TO BE PLOTTED
0006      NUMPAT= PATTERN NUMBER SPECIFIED BY USER
0007      NASH= CONTOURS BELOW NASH ARE PLOTTED AS DASHED LINES
0008
0009      DIMENSION A(151,151),RA(151),RR(151),X(151),Y(151)
0010      REAL CONTOUR(10)
0011      COMMON /APR.Y/A
0012      CALL PLOT(RA,0.,23)
0013      CALL FACTOR (0.7)
0014      JJ=1
0015      MS=M
0016      NS=N
0017      PATIN=MS/NS
0018      SCALE=10.
0019      ANM=AMAX0(N-1,M-1)
0020      IF (PATIN-1.0)1,1,2
0021      1 SX=ANM
0022      SY=PATIN*ANM
0023      GO TO 3
0024      2 SY=1./PATIN*ANM
0025      SY=ANM
0026      3 SMAX=AMAX1(SX,SY)
0027      SS= SX/SMAX
0028      SYS= SY/SMAX
0029      CALL PLOT(2.0,.25,3)
0030      DO 300 I=1,10
0031      CALL PLOT(FLOAT(I)+2.0,.25,?)
0032      CALL PLOT(FLOAT(I)+2.0,.5,?)
0033      CALL PLOT(FLOAT(I)+2.0,.25,2)
0034      CONTINUE
0035      DO 301 I=1,10
0036      YY=FLOAT(I)+.25
0037      CALL PLOT(12.,YY,?)
0038
00000240
00000310
00000335
00000340
00000350
00000360
00000380
00000390
00000400
00000410
00000420
00000430
00000440
00000450
00000460
00000470

```





```

COS 50070AV IV 360N-E0-479 3-9          PLOT2          DATE 03/02/77          TIME          10.01.01
0102      52 IF(RS-RR(J)) 60,60,54
0103      53 RS=RA(J-1)
0104      XS=X (J-1)
0105      YS=Y(K-1)
0106      GO TO 52
0107      54 RS=RR(J)
0108      XS=X (J)
0109      YS=Y (K)
0110      GO TO 60
0111      60 RM=RS
0112      XM=XX
0113      YM=YY
0114      IF(RM-RS) 62, 62,61
0115      61 IF(RM-RL)70 ,62 ,62
0116      62 RM=RA(J-1)
0117      XM=X (J-1)
0118      YM=Y (K-1)
0119      IF(RM-RS) 64,64,63
0120      63 IF(RM-RL) 70,64,64
0121      64 RM = RR(J)
0122      XM=X (J)
0123      YM=Y (K)
0124      YCS=Y5*YC0NA
0125      YCM=YM*YC0NA
0126      YCL=YL*YC0NA
0127      YS=Y5-SY
0128      YM=YM-SY
0129      YL=YL-SY
0130      XCS=XS/SMAX
0131      XCM=XM/SMAX
0132      XCL=XL/SMAX
0133      JJ=1
0134      80 IF(JJ .GT. NCON) GO TO 110
0135      RC=CONTIP(JJ)
0136      IF ( RC .NE. RM ) GO TO 91
0137      IF ( RM .NE. RS ) GO TO 91
0138      IF ( RL .EQ. RM ) GO TO 100
0139      91 IF(RC-PS)100,95,92
00001090
00001100
00001110
00001120
00001130
00001140
00001150
00001160
00001170
00001180
00001190
00001200
00001210
00001220
00001230
00001240
00001250
00001260
00001270
00001280
00001290
00001300
00001310
00001320
00001330
00001340
00001350
00001360
00001370
00001380
00001390

00001420
00001430
00001440
00001450

```





```

0173      DOS PROGRAM IV 360N-F0-470 3-9      PLOT2      DATE      03/02/77      TIME      10.01.01
0179      VY =Y (K)
0180      GO TO 37
0181      118 CONTINUE
0182      CALL FACTOR(1.0)
0183      CALL PLOT(SCALE+6.,0.,23)
0184      CALL ZIP(3,3,18)
0185      RETURN
          END
00001820
00001830
00001840
          00001860
          00001870

```

REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR

SUBROUTINE PLOT3

PURPOSE: TO DRAW A PERSPECTIVE VIEW OF A CONTOURED SURFACE.

DESCRIPTION OF PARAMETERS AND IMPORTANT VARIABLES:

N -- NUMBER OF DATA POINTS ALONG FIRST AXIS.

M -- NUMBER OF DATA POINTS ALONG THE SECOND AXIS.

NUMPAT -- PATTERN NUMBER (FOR LABELLING).

K -- CODE THAT TELLS WHETHER TO DRAW THE GRID LINES:

K=1: ALONG THE M-DIMENSION ONLY.

K=2: ALONG THE N-DIMENSION ONLY.

K=3: ALONG BOTH DIMENSIONS.

DISTS -- DISTANCE FROM SURFACE TO EYE WHEN PERSPECTIVE IS CALCULATED -- SDISTS > 6 USUALLY WON'T SHOW ANY DISTORTION DUE TO PARALLAX.

YAW -- (IN DEGREES) HOW FAR THE OBJECT IS TURNED AWAY FROM THE VIEWER.

PITCH -- (IN DEGREES) HOW THE SURFACE IS LOWERED OR RAISED AT THE FRONT EDGE. (POSITIVE PITCH TENDS TO EXPOSE THE TOP OF THE FIGURE).

SIZE -- (IN INCHES) THE SIZE OF THE CURVE THAT ENCLOSES THE FIGURE.

KODE -- "HIDDEN LINE" SWITCH. IF KODE=0 DO NOT DRAW HIDDEN LINES...IF KODE=1, ALL HIDDEN LINES ARE PLOTTED.

MGN -- WHETHER TO DRAW THE OUTLINE OF THE CURVE TO HELP ORIENT THE VIEWER. MGN=0: DO NOT DRAW ANY OUTLINE OF THE CURVE. MGN=1: DRAW THE OUTLINE OF THE CURVE SEPARATE

FROM THE FIGURE. MGN=2: DRAW THE OUTLINE OF THE CURVE SUPERIMPOSED ON THE SURFACE PLOT. MGN=3: DRAW ONLY THE THREE EDGES OF THE CURVE THAT MEET AT THE ORIGIN, SUPERIMPOSED ON THE SURFACE PLOT.

SCALE - HOW TALL TO MAKE THE SURFACE RELATIVE TO THE HEIGHT OF THE CURVE. SCALE=0: DO NOT SCALE THE DATA AT ALL BUT TRUST THE USER THAT THE DATA IS NOT SO HIGH THAT IT RUNS OFF THE PAPER. SCALE=1: SCALE THE DATA SO THE TOP OF THE DATA JUST TOUCHES THE TOP OF THE CURVE. SCALE=0.3: SCALE THE DATA SO THE TOP OF THE SURFACE IS THREE TENTHS AS HIGH AS THE CURVE.

TASKS.

I. IT IS VERY EXPENSIVE TO DRAW OPAQUE SURFACES, BECAUSE THE PROGRAM HAS TO DETERMINE THE VISIBILITY OF EVERY POINT, THE COMPUTER TIME DOUBLES OR TRIPLES...DEPENDING ON HOW MANY LINE SEGMENTS ARE PARTIALLY VISIBLE.

II. THE CONTENTS OF ARRAY A ARE DESTROYED IN COMPUTATION.

COMMON BLOCKS REQUIRED:

COMMON /ARRAY/ A  
 COMMON /THREE6/ ANGA,ANGR,HV,D,SH,SV  
 COMMON /THREE7/ SL,SM,SN,CX,CY,CZ,OX,OY,OZ,SD

SUBROUTINE AND FUNCTION SUBPROGRAMS REQUIRED:

THREE2  
 THREE3  
 THREE4  
 THREE5  
 PLOT

10.01.24

TIME

03/02/77

DATE

MAINPGM

3-9

360N-FA-479

IV FORTRAN

FACTOR  
SYMBOL  
NUMBER

REFERENCE: HOWARD JESPERSON, IOWA STATE UNIVERSITY.

MODIFIED FOR USE AT VPI BY: ROBERT D. KEPHART.

S. R. KAUFEMAN

W. L. STUTZMAN

F. L. COFFEY

SUBROUTINE PLOT3(N,M,NUMPAT)

COMMON /ARRAY/ A

COMMON /THREE6/ ANGA, ANGR, HV, D, SH, SV

COMMON /THREE7/ SL, SM, SN, CX, CY, CZ, OX, OY, OZ, SD

COMMON /DIMENSION H(10), V(10), X(2), Y(2), Z(2), XP(8), A(151, 151)

K=3

STATS=6.0

PITCH=30.

YAW=45.

SIZE=10.

KORD=0

YCN=0

SCALE=1.0

CALL FACTOR(1.0)

CALL PLOT(8., 2, 23)

CALL PLOT(.4, 0., 2)

CALL PLOT(.4, 8., 2)

CALL PLOT(0., 8., 2)

CALL PLOT(0., 0., 2)

CALL ZIP(3, 3, 18)

CALL SYMBOL(0.3, 1.0, 0.12, 10HPATTERN = ,90., 10)

00000320

00000330

00000370

00000389

00000390

00000400

00000410

00000420

00000430

00000440

00000450

00000470

00000480

00000510

00000520

00000530

00000540

0001

0002

0003

0004

0005

0006

0007

0008

0009

0010

0011

0012

0013

0014

0015

0016

0017

0018

0019

0020

0021

0005 FORTRAN IV 360N-F0-479 3-9 PLOTS DATE 03/02/77 TIME 10.01.24

```

0022      FNUM=FLOAT(NUMPAT)
0023      CALL NUMBER(0.3,2.130 ,0.12,FNUM,90.,-1)
0024      CALL ZIP(1,3,18)
0025      CALL PLOT(1.5,-.2,23)
0026      C*****
0027      ANGA = (YAW+270.) * .0174532
0028      ANGR = PITCH * .0174532
0029      HV = SIZE
0030      C DIRECTION COMPONENTS TO THE EYE.
0031      SL = -COS( ANGA ) * COS( ANGR )
0032      SM = -SIN( ANGA ) * COS( ANGR )
0033      SN = -SIN( ANGR )
0034      IF ( ABS( SN ) .NE. 1.0 ) GO TO 10
0035      WRITE( 6 , 20 )
0036      20  FORMAT ( '1', 20X, 20(' '), / '0', 'YOU ARE ATTEMPTING TO LOOK
0037      :K STRAIGHT DOWN ( OR UP ) AT THE SURFACE ' )
0038      GO TO 2150
0039      10  CONTINUE
0040      SN = 1.0 / SQRT( 1.0 - SN ** 2 )
0041      X(1) = 1
0042      X(2) = N
0043      Y(1) = 1
0044      Y(2) = M
0045      T=MAX0(M,N)
0046      C FIND THE DIAGONAL OF THE "CURE".
0047      D = M ** 2 + N ** 2 + T ** 2
0048      D = SQRT ( D )
0049      SCL = DISTS * D
0050      C COORDINATES OF YOUR EYE.
0051      CY = -SL * SCL
0052      CW = -SM * SCL
0053      CZ = -SN * SCL
0054      C COORDINATES OF THE PROJECTION PLANE.
0055      CX = CY + D * SL
0056      CY = CY + D * SM
0057      CZ = CZ + D * SN
0058      C *****
0059      GO TO 2150

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0053      C      WRITE(6,100) CX,CY,CZ
0054      C      WRITE(6,100) QX,QY,QZ
0055      100      FORMAT(1X,3F15.3)
0056      2050 Z(2)=A(1,1)
0057      Z(1)=A(1,1)
0058      DO 1000 J=1,N
0059      DO 1000 K=1,M
0060      Z(1)=AMINI(Z(1),A(J,K))
0061      Z(2)=AMAX1(Z(2),A(J,K))
0062      1000 CONTINUE
0063      RANGE= (Z(2)-Z(1))
0064      DDL=1.0
0065      IF(SCALF.NE.0) DDL=T/RANGE*SCALE
0066      C      SCALE THE SURFACE TO MAKE A "CURE".
0067      DO 30 I = 1 , N
0068      DO 30 J = 1 , M
0069      A ( I , J ) = ( A ( I , J ) - Z ( I ) ) * DDL
0070      30 CONTINUE
0071      Z(1) = 0.0
0072      Z(2) = T
0073      2080 CALL THREE2 ( X, Y, Z, XP, H, V, KDPF)
0074      DO 2130 I = 1, 8
0075      H( I ) = ( ( XP(I) - QX ) * SM - ( H(I) - QY ) * SL ) * SD
0076      V ( I ) = ( V( I ) - QZ ) * SD
0077      2130 CONTINUE
0078      2100 H(10)=H(1)
0079      H(9)=H(1)
0080      DO 1001 J=1,8
0081      H(9)=AMINI(H(9),H(J))
0082      H(10)=AMAX1(H(10),H(J))
0083      1001 CONTINUE
0084      2120 V(9)=V(1)
0085      V(10)=V(1)
0086      DO 1002 J=1,8
0087      V(9)=AMINI(V(9),V(J))
0088      V(10)=AMAX1(V(10),V(J))
0089      1002 CONTINUE
0090      15( MGN ,EQ. 0) GO TO 2140
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008 FORTRAN IV 360N-EO-479 3-9 PLOTS DATE 03/02/77 TIME 10.01.74

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0088 S=HV
0089 IF(MGN.EQ.1) S=1.5
0090 SH = S/ (H(10)-H(9) )
0091 SV = S/ (V(10)-V(9) )
0092 SH = SIGN( AMINI(SH,SV), SH )
0093 SV = SIGN(SH,SV)
0094 IF(MGN.EQ.1)CALL PLOT (0.,2.,-3)
0095 CALL SYMBOL((H(1)-H(9))*SH,(V(1)-V(9))*SV,.14,'0',0.,1)
0096 CALL SYMBOL((H(3)-H(9))*SH,(V(3)-V(9))*SV,.14,'M',0.,1)
0097 CALL SYMBOL((H(2)-H(9))*SH,(V(2)-V(9))*SV,.14,'Z',0.,1)
0098 CALL SYMBOL((H(5)-H(9))*SH,(V(5)-V(9))*SV,.14,'N',0.,1)
0099 CALL PLOT(.03,.05,-3)
0100 CALL PLOT ( (H(1)-H(9))*SH, (V(1)-V(9))*SV,3)
0101 CALL PLOT ( (H(2)-H(9))*SH, (V(2)-V(9))*SV,2)
0102 CALL PLOT ( (H(1)-H(9))*SH, (V(1)-V(9))*SV,2)
0103 CALL PLOT ( (H(3)-H(9))*SH, (V(3)-V(9))*SV,2)
0104 CALL PLOT ( (H(1)-H(9))*SH, (V(1)-V(9))*SV,2)
0105 CALL PLOT ( (H(5)-H(9))*SH, (V(5)-V(9))*SV,2)
0106 IF(MGN.EQ.3) GO TO 2139
0107 CALL PLOT ( (H(6)-H(9))*SH, (V(6)-V(9))*SV,2)
0108 CALL PLOT ( (H(2)-H(9))*SH, (V(2)-V(9))*SV,2)
0109 CALL PLOT ( (H(4)-H(9))*SH, (V(4)-V(9))*SV,2)
0110 CALL PLOT ( (H(3)-H(9))*SH, (V(3)-V(9))*SV,2)
0111 CALL PLOT ( (H(7)-H(9))*SH, (V(7)-V(9))*SV,2)
0112 CALL PLOT ( (H(5)-H(9))*SH, (V(5)-V(9))*SV,2)
0113 CALL PLOT ( (H(6)-H(9))*SH, (V(6)-V(9))*SV,2)
0114 CALL PLOT ( (H(8)-H(9))*SH, (V(8)-V(9))*SV,2)
0115 CALL PLOT ( (H(4)-H(9))*SH, (V(4)-V(9))*SV,2)
0116 CALL PLOT ( (H(9)-H(9))*SH, (V(8)-V(9))*SV,2)
0117 CALL PLOT ( (H(7)-H(9))*SH, (V(7)-V(9))*SV,2)
0118 IF(MGN.EQ.1) GO TO 2140
0119 CALL PLOT (AINT((H(10)-H(9))*SH+2.),-2.05,23)
0120 CALL THREE3(X,Y,N,M,H,V,K,KODE)
0121 CONTINUE
0122 LL PLOT(16.,0.,23)
0123 CALL ZIP(3,3,18)
0124 PFTURN
0125 END

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REPRODUCIBILITY OF THE ORIGINAL PAGE IS POOR



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0001      SUBROUTINE THREE2 ( I, Y, Z, XP, H, V, KODE)
0002      C FIND THE CORNERS OF THE ROTATED CUBE.
0003      C
0004      DIMENSION X(2), V(2), Z(2), H(10), V(10), XP(8)
0005      C
0006      050  L = 0
0007      070  DO 180 I = 1, 2
0008      C
0009      090  DO 170 J = 1, 2
0010      C
0011      110  DO 160 K = 1, 2
0012      C
0013      130  L = L + 1
0014      140  CALL THREE4 ( X(I), Y(J), Z(K), XP( L ),
0015      1      H(L), V( L ), KODE )
0016      160 CONTINUE
0017      170 CONTINUE
0018      180 CONTINUE
0019      190 RETURN
0020      END

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10.01.51

TIME

DATE 03/02/77

THREE2

360N-EC-479 3-9

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0001      SURROUTINE THREE4, ( X, Y, Z, XP, YP, ZP, KODE;
0002      C  FIND THE LOCATION OF A POINT IN THE ROTATED CURF.
0003      COMMON /THREE6/ ANGA, ANGB, HV, D, SH, SV
0004      COMMON /THREE7/SL, SM, SN, CX, CY, CZ, QX, QY, QZ, SD
0005      SK = 0 / ( (X - CX) * SL + (Y - CY) * SM + (Z - CZ) * SN)
0006      XP = CX + SK * (X - CX)
0007      YP = CY + SK * (Y - CY)
0008      ZP = CZ + SK * (Z - CZ)
0009      RETURN
0010      END

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DCS FORTRAN IV 360N-FC-479 3-9      THREE4      DATE 03/02/77      TIME      10.02.00

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0001      C SURROUTINE THREE (X,Y,N,M,H,V,K,KODE)
0002      C DRAW THE FIGURE.
0003      C COMMON /THREE6/ ANGA , ANGB , HV , D , SH , SV
0004      C COMMON /THREE7/SI , SM , SN , CX , CY , CZ , QX , QY , OZ , SD
0005      C DIMENSION X(2),Y(2),H(10),V(10),A(151,151)
0006      C COMMON /ARRAY/ A
0007      C INTEGER UP , DOWN , PFN , P , Q
0008      C INTEGER PI , PO
0009      C
0010      C CAN USE I / 32 OR I / 64 FOR FINEP INTERPOLATION
0011      C
0012      C UP = 3
0013      C DOWN = 2
0014      C SH = HV / ( H ( 10 ) - H ( 9 ) )
0015      C SV = HV / ( V ( 10 ) - V ( 9 ) )
0016      C SH = SIGN(AMINI(SH,SV),SH)
0017      C SV = SIGN(SH,SV)
0018      C MM = M
0019      C NV = N
0020      C ORO I=(K-1) 100,120,100
0021      C I 100 I=(K-3) 110,120,110
0022      C
0023      C DRAW LINES ALONG THE Y-AXIS
0024      C 120 CONTINUE
0025      C L = 0
0026      C LN = 1
0027      C PD = 0.5 * LD
0028      C
0029      C 140 DO 1060 J = 1, M
0030      C Q = 0
0031      C YJ = J
0032      C 160 DO 1030 I = 1, NN
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0001 10.02.09  
0002 00001970  
0003 00001980  
0004 00001990  
0005 00002000  
0006 00002010  
0007 00002020  
0008 00002025  
0009 00002030  
0010 00002040  
0011 00002050  
0012 00002070  
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0015 00002100  
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0017 00002120  
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0025 00002210  
0026 00002230  
0027 00002240  
0028 00002250  
0029 00002260  
0030 00002270  
0031 00002280  
0032 00002290  
0033 00002300  
0034 00002310  
0035 00002320  
0036 00002330  
0037 00002340

03/02/77  
THREE3  
DATE

005 E02TRAN IV 360N-E0-479 3-9 THREE3 DATE 03/02/77 TIME 10.02.09

```

0025 L = L + LD
0026 XI = L
0027 CALL THREE5 ( XI ,YJ , N , M , P ,K00F)
0028 PEN = IIP
0029 IF ( P ) 510 , 520 , 530
0030 CONTINUE
0031 IF ( Q ) 540 , 550 , 540
0032 CONTINUE
0033 IF ( Q ) 610 , 1020 , 610
0034 CONTINUE
0035 IF ( Q ) 540 , 550 , 540
0036 CONTINUE
0037 PEN = DOWN
0038 GO TO 170
0039 CONTINUE
0040 IF ( Y -E0. 1 ) GO TO 170
0041 DI = 00
0042 TN = L - LD
0043 T = TC + DI
0044 PI = )
0045 IF ABS( DI ) -LT. END ) GO TO 570
0046 CALL THREE5 (T,YJ,N,M,P,K00F)
0047 DI = DI * 0.5
0048 IF ( PC -E0. 0 ) GO TO 565
0049 TN = T
0050 PI = PC
0051 T = T - DI
0052 GO TO 560
0053 T = T + DI
0054 GO TO 560
0055 CONTINUE
0056 T = TN
0057 IF ( PI * P ) 170 , 170 , 580
0058 CONTINUE
0059 CONTINUE
0060 ZP = A(L-L0,J)+(T-L+LD)*(A(L,J)-A(L-L0,J))/LD
0061 CALL THREE4(T,YJ,ZP,XP,HH,VV,K00F)
0062 HH = ( ( XP-OX)*SM- (HH - OY )*SL ) * S0

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005 FORTRAN IV 360N-FC-479 3--9          THREE3          DATE 03/02/77          TIME 10.02.09
0063      VV = ( VV - QZ ) * SD
0064      HH = ( HH - H(9) ) * SH
0065      VV = ( VV - V(9) ) * SV
0066      CALL PLOT ( HH , VV , PEN )
0067      PEN = 5 - PEN
0068      GO TO 170
0069      610 CONTINUE
0070      PEN = DOWN
0071      DI = DN
0072      TO = L - LD
0073      T = TO + DI
0074      PI = Q
0075      IF ( ABS( DI ) .LT. ENN ) GO TO 630
0076      CALL THREE5 ( T, YJ, N, M, PO, KODE )
0077      DI = DI * 0.5
0078      IF ( PO .EQ. 0 ) GO TO 625
0079      TO = T
0080      PI = PO
0081      T = T + DI
0082      GO TO 620
0083      T = T - DI
0084      GO TO 620
0085      630 CONTINUE
0086      T = TO
0087      IF ( PI * Q ) 600 , 600 , 590
0088      CALL THREE4 ( XT , YJ , A( L, J ), XP , HH , VV , KODF )
0089      VV = ( VV - QZ ) * SD
0090      HH = ( ( XP-QX)*SM- (HH - QY)*SL ) * SD
0091      HH = ( HH - H(9) ) * SH
0092      VV = ( VV - V(9) ) * SV
0093      CALL PLOT ( HH , VV , PEN )
0094      1020  O = P
C
C
0095      L = L + LD
0096      CONTINUE
0097      LD = -LD
0098      DO = -DO

```

```

0099      COS FORTAN IV 360N-FO-479 3-9      THREFF3      DATE 03/02/77      TIME 10.02.09
C 1060 CONTINUE
C
C 1090 IF(K-3) 2060,1110,2060
C
C DRAW LINES ALONG THE X-AXIS.
C 1110 CONTINUE
C
C L = 0
C LD = 1
C PD = 0.5 * LD
C 1140 DO 2040 I = 1 , N
C XI = I
C Q = 0
C 1160 DO 2020 J = 1 , MM
C L = L + LD.
C YJ = L
C CALL THREFF5 (XI,YJ,N,M,P,KODE)
C PFN = IIP
C 1510 CONTINUE
C IF ( P ) 1510 , 1520 , 1530
C 1520 CONTINUE
C IF ( Q ) 1540 , 1550 , 1540
C 1530 CONTINUE
C IF ( Q ) 1610 , 2010 , 1610
C 1540 CONTINUE
C IF ( Q ) 1540 , 1550 , 1540
C PFN = PDWN
C GO TO 1170
C 1550 CONTINUE
C IF ( J .EQ. 1 ) GO TO 1170
C PD = PD
C PD=L-LD
C T = TC + PI
C PI = Q
C 1560 IF ( ABS( PI ) .LT. FND ) GO TO 1570
C CALL THREFF5 (XI,T,N,M,PD,KODE)

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00003470

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0130      OI = OI * 0.5
0131      IF ( PO .EQ. 0 ) GO TO 1565
0132      TO = T
0133      PI = PO
0134      T = T - OI
0135      GO TO 1560
0136      T = T + OI
0137      GO TO 1560
0138      CONTINUE
0139      T = TO
0140      IF ( PI * P ) 1170 , 1170 , 1580
0141      CONTINUE
0142      CONTINUE
0143      ZP=A(I,L-LD) + (T-L+LD) * (A(I,L) - A(I,L-LD))/LD
0144      CALL THRFF4 ( XI , T , ZP , XP,HH,VV ,KODE)
0145      HH = ( ( XP-OX)*SM- (HH - OY )*SL ) * SO
0146      VV = ( VV - OZ ) * SO
0147      HH = ( HH - H(9) ) * SH
0148      VV = ( VV - V(9) ) * SV
0149      CALL PIOT ( HH , VV , PEN )
0150      PEN = 5 - PEN
0151      GO TO 1170
0152      CONTINUE
0153      OFN = OFNM
0154      OI = OO
0155      TO = L - LD
0156      T = TO + OI
0157      PI = O
0158      IF ( ABS(OI) .LT. FND ) GO TO 1630
0159      CALL THRFF5 ( XI,T,M,PQ,KODE)
0160      OI = OI * 0.5
0161      IF ( PO .EQ. 0 ) GO TO 1625
0162      TO = T
0163      PI = PO
0164      T = T + OI
0165      GO TO 1620
0166      T = T - OI
0167      GO TO 1620

```







C COMMON /THREE7/ SL,SM,SN,CX,CY,CZ,QX,QY,QZ,SD  
 C  
 C  
 C SURROUTINE AND FUNCTION SUBPROGRAMS REQUIRED:NONE.  
 C  
 C

```

0001 SURROUTINE THREE5 (XI,YJ,M,N,P,KONE)
0002 DIMENSION Z(151,151)
0003 COMMON /THREE6/ ANGA, ANGR, HV, D, SH,SV
0004 COMMON /THREE7/SL,SM,SN,CX,CY,CZ,QX,QY,QZ,SD
0005 COMMON /ARRAY/ Z
0006 INTEGER CUM, CNT, P
0007 REAL I, J, IT, JJ
0008 IF( KODE .EQ. 1) GO TO 78
0009 IP = XI
0010 JC = YJ
0011 ZR = Z ( IR, JC )
0012 IF ( XI .EQ. IP ) GO TO 2
0013 ZR = Z( IR, JC ) + ( XI - IR ) * ( Z( IR + 1, JC ) - Z( IR, JC ) )
0014 GO TO 4
0015 2 IF ( YJ .EQ. JC ) GO TO 4
0016 ZR = Z( IP, JC ) + ( YJ - JC ) * ( Z( IP, JC + 1 ) - Z( IP, JC ) )
0017 4 CONTINUE
0018 XEND = 0.0
0019 CX = 0.0
0020 YMULT = 0.0
0021 ZMULT = 0.0
0022 IF (XI .EQ. CX ) GO TO 10
0023 YMULT = (YJ - CY ) / ( XI - CX )
0024 ZMULT = ( ZR - CZ ) / ( XI - CX )
0025 CX = 1.0
0026 XEND = M + 1
0027 IF ( XI .LT. CX ) GO TO 10
0028 CX = -1.0
0029 XEND = 0.0
0030 CONTINUE
0031 YEND = 0.0
    
```

```

0032      DV = 0.0
0033      XMULT = 0.0
0034      IF ( YJ .EQ. CY ) GO TO 20
0035      XMULT = ( XI - CX ) / ( YJ - CY )
0036      IF ( ZMULT .EQ. 0.0 ) ZMULT=(ZB - CZ ) / ( YJ - CY )
0037      DV = 1.0
0038      YEND = N + 1
0039      IF ( YJ .LT. CY ) GO TO 20
0040      DV = -1.0
0041      YEND = 0.0
0042      20 CONTINUE
0043      CNT = 0
0044      CNT = 0
0045      P = 0
0046      XR = XI
0047      YR = YJ
0048      30 CONTINUE
0049      II = AINT( XR )
0050      JJ = AINT( YR )
0051      XSTEP = DX
0052      YSTEP = DY
0053      IF ( XR .EQ. II ) GO TO 40
0054      IF ( DX .LT. 0.0 ) XSTEP = 0.0
0055      GO TO 45
0056      40 IF ( YR .EQ. JJ ) GO TO 45
0057      IF ( DY .LT. 0.0 ) YSTEP = 0.0
0058      45 CONTINUE
0059      I = II + XSTEP
0060      J = JJ + YSTEP
0061      IF ( I .EQ. XEND ) GO TO 80
0062      IF ( J .EQ. YEND ) GO TO 80
0063      XR = CX + XMULT * ( J - CY )
0064      YR = CY + YMULT * ( I - CX )
0065      IF ( CX .LT. 0.0 ) GO TO 55
0066      IF ( XR .LT. I ) GO TO 60
0067      XR = I
0068      GO TO 65
0069      55 IF ( XR .LT. I ) GO TO 50
0070      60
0071      65
0072      70
0073      75
0074      80
0075      85
0076      90
0077      95
0078      100
0079      105
0080      110
0081      115
0082      120
0083      125
0084      130
0085      135
0086      140
0087      145
0088      150
0089      155
0090      160
0091      165
0092      170
0093      175
0094      180
0095      185
0096      190
0097      195
0098      200
0099      205
0100      210
0101      215
0102      220
0103      225
0104      230
0105      235
0106      240
0107      245
0108      250
0109      255
0110      260
0111      265
0112      270
0113      275
0114      280
0115      285
0116      290
0117      295
0118      300
0119      305
0120      310
0121      315
0122      320
0123      325
0124      330
0125      335
0126      340
0127      345
0128      350
0129      355
0130      360
0131      365
0132      370
0133      375
0134      380
0135      385
0136      390
0137      395
0138      400
0139      405
0140      410
0141      415
0142      420
0143      425
0144      430
0145      435
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0151      465
0152      470
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0154      480
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0156      490
0157      495
0158      500
0159      505
0160      510
0161      515
0162      520
0163      525
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0176      590
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0178      600
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0181      615
0182      620
0183      625
0184      630
0185      635
0186      640
0187      645
0188      650
0189      655
0190      660
0191      665
0192      670
0193      675
0194      680
0195      685
0196      690
0197      695
0198      700
0199      705
0200      710
0201      715
0202      720
0203      725
0204      730
0205      735
0206      740
0207      745
0208      750
0209      755
0210      760
0211      765
0212      770
0213      775
0214      780
0215      785
0216      790
0217      795
0218      800
0219      805
0220      810
0221      815
0222      820
0223      825
0224      830
0225      835
0226      840
0227      845
0228      850
0229      855
0230      860
0231      865
0232      870
0233      875
0234      880
0235      885
0236      890
0237      895
0238      900
0239      905
0240      910
0241      915
0242      920
0243      925
0244      930
0245      935
0246      940
0247      945
0248      950
0249      955
0250      960
0251      965
0252      970
0253      975
0254      980
0255      985
0256      990
0257      995
0258      1000

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0070      DO5 FORTRAN IV 360N-EN-479 3-9      THREES      DATE 03/02/77      TIME 10.02.30
0071      60 Y8 = J
0072      65 CONTINUE
0073      Z8 = CZ + ZMULT * ( XB - CX )
0074      IP = I
0075      JC = J
0076      IF ( Y8 .NE. J ) GO TO 70
0077      IPX = I - DX
0078      Z5 = Z( IP, JC ) - DX * ( XB - I ) * ( Z( IPX, JC ) - Z( IR, JC ) )
0079      GO TO 75
0080      JDY=J-DY
0081      Z5 = Z( IP, JC ) - DY * ( Y8-J ) * ( Z( IR, JDY ) - Z( IR, JC ) )
0082      75 CONTINUE
0083      SGN = I
0084      IF ( 78 .LT. Z5 ) SGN = -I
0085      CUM = CUM + SGN
0086      CNT = CNT + I
0087      IF ( IARS ( CUM ) .EQ. CNT ) GO TO 30
0088      GO TO 90
0089      P=I
0090      GO TO 95
0091      80 CONTINUE
0092      P = I
0093      IF ( CUM ) 84 , 86 , 90
0094      84 P = -I
0095      GO TO 90
0096      86 CONTINUE
0097      IF ( Z8 .LE. CZ ) GO TO 90
0098      P = -I
0099      90 CONTINUE
0100      95 RETURN
      END

```