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# **ACM Technical Report**

**ACM-TR-120** 

NASA CR- $16055.$ 

**(NASA-CR-160552) CODING, TESTING AND 880-20289 DOCOlBlTATIOI OF PROCESSORS FOB TEE PLI68T DBSIGY SYSTEM Final Report (Analytical and Computational Mathematics, Inc.) 24 p Phono Buclas HC A02/LIP A01 CSCL 22A C3/13 47542** 

> **Coding, Testing and Documentation of Processors for the Flight Design System**

> > **FINAL REPORT Contract NASS-15787**

ANALYTICAL AND COMPUTATIONAL MATHEMATICS,



**COD1 NG, TESTING AND DOCUMENTATION OF PROCESSORS FOR THE FLIGHT DESIGN SYSTEM** 

**FINAL REPORT** 

**ANALYTICAL AND COMPUTATIONAL MATHEMATICS, INC. 1275 SPACE PARK DRIVE, SUITE 114 HOUSTON, TEXAS 77058 FEBRUARY 22, 1980** 

**This report was prepared for the NASAIJohnson Space Center under Contract NAS9-15787.** 

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Coding, Testing and Documentation of Processors For the Flight Design System

#### 1.0 Introduction

This is the final summary report for the work carried out under Contract **NAS9-15787.** The purpose of this report is to give an overview of the work. The individual tasks are summarized in Section 2 through Section 8. A statement on New Technology is given in Section 9. Appendix **A** gives a comprehensive list of all the items delivered under this contract. Appendix B lists the source code card count for all delivered routines.

#### **2.0** Familiarization and Numerical Accuracy Investigations

All ACM personnel assigned to the FDS work were required to become familiar with the Interdata 8/32 Computer system. A key **ACM** person was sent to a two week course on the computer system.

A study was made on the accuracy of single and double precision arithmetic, and mathematical functions. It was determined that all orbit propagation routines should be coded in double precision. This standard was implemented in all propagation routines and related routines. Some exceptions were made with perturbation model routines where theoretical considerations implied that single precisioa arithmetic would provide sufficient accuracy.

### 3.0 Design and Implementation of Basetime Initialization Processor

This program (BASTM) was the first complete processor delivered by **ACM.** It is needed in order to execute other application processors.

The Basetime Processor (BASTM) provides the capability to establish the basetime (epoch) for a given session and thereby initialize the base date array IBDATE . Two reference times, Mission Elapsed Time (MET) and Phase Elapsed Time (PET), can be generated optionally within the !BDATE array. It also, at the user's option, initializes solar and lunar coefficient arrays that are required by processors using analytical solar and lunar ephemeris routines. All above mentioned optional output can be generated without updating the entire IBDATE array.

#### 4.0 Design and Implementation of Utility Routines

The following utility routines were developed and delivered,

- (a) SUN Given the time from base date, SUN computes the position of the sun using analytical conic expressions. SUN will output the position in two forms: in cartesian coordinates and also in position magnitude, right ascension, declination. Both are referenced to the TEG system.
- (b) MOON Given the time from base date, routine MOON analytically computes the position of the moon by using a truncated for of Brown's series expansions. The position will be output in two forms: Cartesian coordinates and also radius magnitude, decljnation, right ascension. Both forms are referenced to the TEG system defined by previous execution of BASTM processor.
- (c) TOUR This set of routines allows any of the orbit propagation processors (except ORBTIM) to stop on any one of the following fifteen stop options:
	- 1) Time
	- 2) Delta time
	- 3) N<sup>th</sup> apsidal crossing after threshold time
	- 4) N<sup>th</sup> apogee crossing after threshold time
	- 5) N<sup>th</sup> perigee crossing after threshold time
- 6) N<sup>th</sup> ascending node after threshold time
- 7) N<sup>th</sup> descending node after threshold time
- 8) Argument of latitude after threshold time
- 9) Delta Orbit Count
- 10) Declination after threshold time
- 11) Longitude after threshold time
- 12) Right ascension after threshold time
- 13) Radius after threshold time
- 14) Altitude, above a spherical earth, after threshold time
- 15) Central angle
- d) CONIC This routine will use the two-body gravity model to propagate a state vector up to a user specified final time.
- e) GCONIC This set of routine will use the two-body gravity model to propagate a state vector until a specified termination parameter reaches a specified value.
- f) RKF45 The utility routine RKF45 will compute the solution of a system of first order ordinary differential equations. The variable step option causes an optimum stepsize to be computed after each step.

### **5.0** Design and Implementation of a Conic Coasting Flight Processor

The Conic Coasting Flight Processor (CCOAST) will analytically propagate a given state vector and terminate when a user-specified termination parameter reaches a user specified value. It will optionally generate intermediate state data on the user specified intervals of time. A two-body earth gravity model is assumed.

### **6.0** Design and Implementation of an Analytical Coasting Flight Processor

The analytical Coasting Flight Processor (ACOAST) will analytically propagate a given state vector and terminate when a user-specified termination parameter reaches a user-specified value. It will optionally generate in'ermediate state data on the user-specified intervals of time. The perturbation model includes a non-spherical earth and atmospheric drag. Sun-moon gravitational perturbations and solar radiation pressure are not included in ACOAST.

ACOAST will propagate a given state vector to the specified termination parmeter. A choice between two propagation modes is available: ASOP or AEG. Both of these modes produce a solution using finite mathematical expressions and will provide various levels of accuracy as determined by the level of precision in the models of the perturbing accelerations. ASOP will provide the higher level of precision, with the capability to include any order or degree geopotential model. In the ASOP mode, the user may choose any of the geopotential models provided in the FDS-2 master data base. In accounting for the atmospheric dray cflccts, both AEG and ASOP will call the user-specified density model; either the 62 standard, Jacchia (1970) or Jwchia **(1971).** 

### **7.0** Design and Implementation of a Precision Coasting Flight Processor

The Precision Coasting Flight Processor **(PCOAST)** will numerically propagate a given state vector and terminate when a user-specified termination parameter reaches a user-specified value. It will optionally generate intermediate state data on the user-specified intervals of time.

PCOAST includes a precision perturbation model, including any combination of the following:

- · an earth gravitation field,
- . sun and moon point mass gravitational forces,
- *e* atmospheric drag,
- *•* sotar radiation pressure.

This program can be used for closed earth-centered orbits, **as** well **as** escape trajectories.

#### 8.0 Design and Implementation of an Orbit Lifetime Processor

The Orbit Lifetime Processor ORBTIM provides the capability to analyze orbital lifetimes of satellites in near earth orbit. It will also optionally generate and display an ephemeris of intermediate states data that may subsequently be used as input to other processors.

The initial state is propagated forward in time until one of the following end conditions is met:

- 1) the specified time interval  $(\Delta \tau)$  is elapsed,
- 2) the orbit perigee falls below a specified reentry altitude, or
- 3) propagation truncation errors become too large.

The trajectory perturbation model includes:

- an earth gravitation field,
- **sun** and moon point mass gravitational forces,

atmospheric drug,

solar radiation pressure.

The propagation is accomplished by using a seminumerical orbit prediction algorithm based on a multirevolution integra**tion technique.** This method makes use of the fact that the orbital motion **or** a vatellite is nearly periodic from **revolu**tion to revolution, as measured from some orbital **refereme**  point such aa perigee. The orbit is propagated ahead one revolution using nuwerical integration with all significant perturbing forces included. This integration provides an accuate determination of the change **of** the orbital elements over one revolution and these changes in turn **are** used to **ex**trapolate the orbit **ahead M** revolutions. This process is repeated until one **of** the specified **end** conditions is reached. The single revolution integration is done with the Runge-Kutta fourth/fifth order routine **RKF45.** 

#### 9.0 Statement on New Technology

The work carried out under this contract consisted of software conversion, integration, documentation and check-out. There were no New Technology items identified as being developed under this contract.

### **APPENDIX A**

#### **ITEMS DELIVERED UNDER THIS CONTRACT**

#### **LIST OF ITEMS DELIVERED UNDER CONTRACT NASO-15787**



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### **APPENDIX B**

### **SOURCE CODE CARD COUNT FOR ALL DELIVERED ROUTINES**

# **Utility Routines**

# **Source Code Card Count**

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# - **BaseTime Processor (BASTM)**

**Source Code Card Coznt** 

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## Conic Coasting Flight Processor (CCOAST) Source Code Card Count





# Analytical Coasting Flight Processor (ACOAST) Source Code Card Count



### **Analytical Coasting Flight Processor (ACOAST)** Source Code Card Count (cont.)

**Contract Management** 

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## Precision Coasting Flight Processor (PCOAST) Source Code Card Count

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## Precision Coasting Flight Processor (PCOAST) Source Code Card Count (cont.)

# Orbit Lifetime Processor (ORBTIM)

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## Source Code Card Count

