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NASA CASE NO. MFS-28524-1
PRINT FIG. #1

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(NASA-Case-MFS-28524-1) THERMALLY ISOLATED
DEPLOYABLE SHIELD FOR SPACECRAFT Patent
Application (NASA) 25 p CSCL 22B

N91-25167

Unclas

G3/18 0024765

THERMALLY ISOLATED DEPLOYABLE SHIELD FOR SPACECRAFT**PATENT APPLICATION ABSTRACT**

The present field of technology of this invention relates generally to shields for spacecrafts. The present invention relates particularly to deployable shields for generally cylindrical spacecraft and to the actuation means utilized with these shields.

Referring to Figure 1 of the application, a plurality of relatively flexible lattice structures 26 generally encircle craft 22 providing 360 degree radial coverage therearound. Lattice structures 26 are capable of flexing to varying uniform curvatures giving shield 20 variable diameter capability. Replaceable shield panels 32 are removably affixed to lattice structures 26, together forming lattice panels 34. A plurality of medial spring actuators 56 and side spring actuators 58 translate lattice panels 34 radially outward a fixed distance 24 from craft 22. Spring actuators 56 and 58 respectively, thermally isolate lattice panels 34 from craft 22 by employing the minimum thermal interfaces between the panels and the craft. Tension bands 68 hold the shield relatively tight against the craft prior to deployment thereof. A plurality of flexure close-out assemblies, which are in turn comprised of a plurality of flexure elements 76, provide light sealing and protection for the annular areas between the deployed shield 20 and craft 22.

The present invention has significant advantages over other shielding schemes for spacecraft, particularly from weight, thermal isolation, reliability, and impingement survivability standpoints. The present invention is suited for spacecraft in general and in particular for cylindrical craft such as space station.

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**TITLE: THERMALLY ISOLATED DEPLOYABLE SHIELD FOR SPACECRAFT
ORIGIN OF THE INVENTION**

The invention described herein was made by employees of the United States Government and may be manufactured and used by or for the Government for governmental purposes without the payment of any royalties therein or therefor.

BACKGROUND OF THE INVENTION

The present invention generally concerns a thermally isolated deployable meteoroid/debris shield for spacecraft in general, and more particularly concerns a weight efficient deployable shield of segments providing 360 degree coverage of a cylindrically shaped spacecraft such as Space Station, while at the same time providing a superior thermal barrier to conductive and radiative heat losses to "cold space" from the spacecraft's internal environment.

Spacecraft destined for long term orbital use must be provided with meteoroid/debris impingement protection. Several factors necessitate this protection: First, the statistical likelihood of meteoroid impingement with long term use; Second, the increasingly larger amounts of orbital space debris released from earth launched spacecraft activity; and Third, the general trend to enhance meteoroid/debris impingement survivability reliability from safety, cost, and weight standpoints.

Additionally, it is an important concern to provide a thermal barrier to internal environment heat loss to "cold space" through conductive and radiative modes, or interface points, existing between the pressurized environment and the meteoroid shield, whether the shield is deployed or fixed. Typically, this has not been as critical of a concern. However, with the advent of Space Station and the opportunity for sustained manned space missions, an effective thermal barrier is a critical requirement. This concern pertains to environmental control system (ECS) sizing, control, power, weight, etc., as well as dew formation, cleanliness, and microbial growth environments. Thus it is important to minimize the interface conductivity through reduction of conduction paths and/or lowering of interface structure/mechanism conductivity, for example through isolation and/or insulation of these interface points.

There are several examples of other methods providing a thermally isolated shielding protection scheme. However, these methods were neither designed for nor adequate for the stringent weight, thermal, and ballistic protection requirements that are so intensive to the thirty year mission of the space station. An example of one such scheme can be seen in the Skylab protection device. Skylab utilized eight individual rigid panels per circumference, or circular cross section. Each individual panel consisted of skin, structure, crank links, prime mover, as well as peripheral elements such as brackets, bearing blocks, light seals, close-outs, skirting, etc. The entire system was retained during ascent with highly loaded, pyro-released tension bands. The system was inherently "heavy" due to the aerodynamic loading and restraint scheme. The eight panel deployment scheme was somewhat complex and possessed a large quantity of components, resulting in reduced reliability of the system.

Another method of providing meteoroid protection is to size the pressurized environment wall thickness to withstand penetration. This method, being extremely heavy, is simply inadequate from a weight sufficiency standpoint for utilization with Space Station.

The use of fixed shields in general is known in the art, but this method is also inadequate. Fixed shields require a large quantity of fastening interfaces resulting in poor thermal characteristics. The fixed shields also exhibit poor ballistic impingement properties.

U.S. Patent No. 4,314,682 to Barnett et al. discloses a spring loaded mechanism for deploying a shield from a space vehicle. The means for deploying the shield includes a plurality of elongated spring members extending outwardly from the body of the space vehicle and a plurality of curved ribs having their ends connected to ends of the extending elongated members. When deployed, the shield is in the general form of an open shell or bathtub-like structure with

end caps at each end, the space vehicle residing within the structure. The shielding material itself is highly flexible radar attenuating material which, prior to deployment, is packed and folded into a jettisonable pod carried along one side of the space vehicle. When the shield is stowed, the spring members are wrapped downwardly around the circumference of the vehicle and held in place by the pod. The shield spontaneously deploys when the pod is jettisoned releasing the spring members which are attached to the shielding material.

U.S. Patent No. 4,919,366 to Cormier discloses a heat resistive wall assembly for space vehicles comprising an inner wall of wrought beryllium or aluminum providing structural support for the vehicle, and an outer wall of interlocking panels of a honeycomb laminate of heat resistive material. An evacuated jacket of insulating material is disposed between the inner and outer walls. The space between the inner and outer walls that is not contained within the evacuated jacket is vented to ambient atmosphere.

U.S. Patent No. 4,578,920 to Bush et al. discloses deployable truss structure having first and second spaced surface truss layers. A passive spring positioned about an elongated shaft serves as the expansion force to move the folded struts from a stowed collapsed position to a deployed operative final truss configuration.

U.S. Patent No. 4,166,598 to Seifert et al. discloses a stowable and inflatable vehicle enshrouding apparatus adapted for use in retaining heat emitted by a large, relatively hot space vehicle. The apparatus includes an inflatable framework external of which is attached a multi-layer superinsulating blanket shroud attached to the inflatable members which comprise the frame. The apparatus is deployed by removing it from stowage and inflating the inflatable support members.

U.S. Patent No. 4,164,339 to McClenny discloses an environmental protection system comprising sheets of thermal insulators superposed one upon the other and deployed over the surface to be protected. A "dead space" thermal insulation, such as a vacuum or simply a high resistance physical separation, is provided between the surfaces. The separation is effected by flaps actuated by an aerodynamic or forced airstream or by the static energy stored in uncoiled flaps. The insulating material is stored on reels and deployed by automatic or manual means.

U.S. Patent No. 4,009,851 to Cable discloses a spacecraft structure having a hollow inner cylindrical member and a plurality of planar bulkheads secured to the outer surface of the inner member and extending radially outward from the inner member. A plurality of planar enclosure panels are secured to the extended edges of the bulkheads and each other to form and enclosed spacecraft structure.

U.S. Patent No. 4,730,797 to Minovitch, U.S. Patent No. 3,547,375 to Mackey, and U.S. Patent No. 3,064,317 to Dobson all relate generally to the field of the present invention.

SUMMARY OF THE INVENTION

An object of the present invention is to provide a thermally isolated deployable shield for spacecraft.

Another object of the present invention is to provide a shield suitable for use with a long term orbital craft.

Yet another object of the present invention is to provide a weight efficient and cost effective shield exhibiting superior thermal barrier capabilities.

Still another object of this invention is to provide a shield exhibiting superior meteoroid and debris impingement protection.

Still another object of this invention is to provide a shield enhancing meteoroid and debris survivability reliability taking into consideration safety, cost, and weight factors.

A further object of the present invention is to provide a shield designed for the stringent weight, thermal, and ballistic protection requirements of the extended life mission of the Space Station.

Another object of the present invention to provide a thermal barrier for a spacecraft to internal heat losses to "cold space."

Yet another object of this invention is to provide means for thermally isolating a deployable shield utilizing minimum shield/craft interfaces.

And still a further object of the present invention is to provide a shield of relatively flexible panels capable of changing curvature thus giving the shield variable diameter capability.

Yet another object of this invention is to provide a deployable shield utilizing flexible structural elements thereby enhancing the shield's weight efficiency.

A further object of this invention is to provide a deployable shield utilizing the minimum number of panels and

peripheral components for weight and reliability considerations.

Another object of the present invention is to provide a deployable shield allowing for shield material thickness downsizing as a result of vessel/shield spacing.

Still another object of the present invention is to provide a more weight efficient shield as the result of the enhanced protection gained from enhanced shield spacing.

A further object of the present invention is to provide a shield end close-out assembly for annular space light sealing and protection utilizing a simplistic design of flexible elements.

Yet another object of the present invention is to provide an improved system for deploying shields from a craft.

These and other objects, aspects, and features of this invention are more particularly discussed and described in the remainder of the specification. Various modifications and alterations to the features, elements, and constructions disclosed herewith may occur to those of ordinary skill in the art, and are intended to come within the spirit and scope of this invention by virtue of present reference thereto. Such modifications and variations may include, but are not limited to, the substitution of functionally equivalent structures and elements for those expressly disclosed, illustrated, or suggested herewith, as well as the interchange of various features and elements previously disclosed. Embodiments of the present invention may also include the elements, and their functional equivalents, disclosed and discussed herein in any combination or standing alone.

Further, it should be understood that, although the present invention is illustrated with and generally discussed in relation with cylindrical craft, this is for ease and clarity of explanation and is not meant to be a

limitation of any kind. The scope and spirit of the present invention is not limited to cylindrical shapes and encompasses, for example, conical or truncated conical shapes such as those found in missile nose cones. The invention may be employed as a protection scheme for any space vehicle or payload in which the launch configuration constrains the maximum envelope required to provide adequate protection for the minimum weight. For example, this invention would be just as suited for any Shuttle or expendable booster launched satellite.

Still further, it is to be understood that different embodiments, as well as different presently preferred embodiments, of this invention may include various combinations of presently disclosed features, or their equivalents (including combinations thereof not expressly shown or stated). In accordance with one exemplary preferred embodiment of the present invention, a thermally isolated variable diameter deployable spacecraft shield is provided which is stowable generally adjacent the craft and deployable to a fixed distance from the craft. The shield generally comprises a plurality of relatively flexible lattice structures capable of flexing to varying curvatures; replaceable shield panels removably attached to the lattice structures, such lattice structure and attached shield panels comprising a lattice panel; securing means for removably securing the shield panels to the lattice structure, the securing means allowing the panels to flex in conjunction with the lattice structure; actuation means for securing the lattice panels relative the craft and for driving the lattice panels a set distance from the craft; and restraining means holding the shield relatively tight against the craft prior to deployment thereof.

In a preferred embodiment of the present invention, the lattice structure comprises a lattice web with relatively rigid longitudinal members and relatively flexible ring

members affixed to the lattice web. The rigid longitudinal members may further comprise an upper longitudinal member and a lower longitudinal member. The lower longitudinal member is preferably utilized along the lattice structure's sides and centerline.

In another preferred embodiment of the present invention, four 90 degree lattice panels are utilized providing 360 radial coverage around the craft. Further, in another preferred embodiment, folding hinge assemblies are disposed between and attached to the lattice panels whereby upon deployment of the shield the hinge assemblies expand between the lattice panels thereby comprising make-up shielding material between the lattice panels and providing continuous 360 radial coverage.

Preferably, the lattice panels are thermally isolated from the craft's structural framework. In one preferred embodiment, this is accomplished by the actuation means comprising a minimum number of thermal interfaces between the craft and shield. These thermal interfaces can preferably include bearing elements having minimum roller/race interface areas.

In another exemplary embodiment according to the present invention, the actuation means include a plurality of spring actuators. More specifically, a set of medial actuators may be provided disposed longitudinally relative the craft and lattice panel along the panel's centerline, preferably along the centerline lower longitudinal member. Preferably, the medial actuators are employed in opposing groups of two to provide a balanced deploying force. Upon deployment, the medial actuators translate the lattice panel outward a fixed distance dictated by the actuators. A set of side actuators are provided disposed generally equally along the lattice panel's sides, preferably along the side lower longitudinal members, and oriented perpendicular to the medial actuators. Upon deployment, the side actuators

locate the edges of the lattice panel radially outward matching the deployed distance of the medial actuators, with the entire panel then having a larger uniform curvature.

5 In another preferred exemplary embodiment according to the present invention, the spring actuators are nonslidably secured to the craft's structural framework and slidably secured to the shield's lattice structure by means of thermally isolating bearing elements.

10 In yet another preferred embodiment in accordance with the present invention, the deployable shield further includes a plurality of flexure close-out assemblies providing light sealing and protection of the annular areas between the deployed shield and the craft's end structure.

15 In accordance with the present invention, apparatus for deploying panels of a craft's variable diameter deployable shield is provided, comprising medial spring actuators oriented longitudinally relative said craft and said panel, said medial actuators being slidably secured to the centerline of said panel and nonslidably secured to said craft's framework so that upon deployment said medial actuators translate said panel outward a fixed distance from said craft; and side actuators oriented perpendicular to said medial spring actuators and located generally equally spaced along said panel's edges, said side actuators slidably secured to said panel's edges and nonslidably secured to said craft's framework so that upon deployment said side actuators locate the edges of said panel radially outward matching the fixed distance of said panel when deployed. The actuators are preferably secured with thermally isolating bearing elements having minimum roller/race interface areas.

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35 In further accordance with the present invention, a flexure close-out assembly is provided for incorporation with a craft's variable diameter deployable shield, said assembly disposed generally at the ends of said shield

between said shield and said craft and capable of flexing
with said shield upon deployment thereof providing annular
space light sealing and protection, comprising a plurality
of flexure elements having cut-out reliefs permitting said
flexure elements to flex with the change in curvature of
said deployable shield; and attaching means, between the
cut-out reliefs, for attaching said flexure elements to said
shield generally at the ends thereof and to said craft's
close-out and light seal assembly.

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BRIEF DESCRIPTION OF THE DRAWINGS

The accompanying drawings, which are incorporated in and constitute a part of this specification, illustrate embodiments of the present invention, including the best mode thereof, and together with the description, serve to explain the principles of the invention.

Figure 1 is a component, exploded perspective view of an exemplary embodiment of the present invention.

Figure 2A illustrates a partial perspective view of an embodiment of the lattice structure according to the present invention.

Figure 2B is a component cross-sectional view of the lattice web and upper longitudinal member of the present invention, taken along line A-A in Figure 1.

Figure 2C is a component cross-sectional view of the lattice web and upper and lower longitudinal members of the present invention, taken along line B-B in Figure 1.

Figure 3 is an enlarged partial perspective view of an exemplary arrangement of the hinge assembly, tension band, and tension bolt assembly shown in an exemplary stowed configuration in accordance with the present invention.

Figure 4 is a schematic cross-sectional view of the shield features of the present invention shown in the stowed configuration, apart from a craft for clarity in illustration.

Figure 5 is an enlarged partial perspective view of the present invention similar to that shown in Figure 3, and depicting the shield in its deployed configuration.

Figure 6 is a schematic cross-sectional view of the shield features of the present invention shown in the deployed configuration, apart from a craft as in Figure 4.

Figure 7A is a partial side view of an exemplary embodiment of the present invention, which may be practiced alone or in combination with other present features,

illustrating the hinge assembly and side spring actuators in the stowed configuration.

5 Figure 7B is a partial side view similar to that of Figure 7 depicting the apparatus in the deployed configuration.

10 Figure 8 is a component perspective view of an embodiment of the present invention, which may be practiced alone or in combination with other present features, depicting apparatus for deploying panels of a deployable shield, particularly illustrating the thermal interfaces and spring prime movers in accordance with the present invention.

15 Figure 9 is a partial component view of an exemplary embodiment of the flexure close-out assembly in accordance with the present invention.

Figure 10 is an enlarged perspective view of the flexure elements in accordance with the present invention.

20 Figure 11 is an enlarged cross-sectional view of an exemplary embodiment of the bushing arrangement of the present invention, taken along the line A-A in present Figure 10.

25 Repeat use of reference characters in the following specification and appended drawings is intended to represent the same or analogous features or elements of the present invention.

DETAILED DESCRIPTION OF THE PREFERRED EMBODIMENTS

Reference will now be made in detail to the present preferred embodiments of the invention, examples of which are illustrated in the accompanying drawings. The following disclosure is for purposes of example only, and is not intended to limit broader aspects of the invention embodied thereby.

Referring in greater detail to the figures, a thermally isolated variable diameter deployable shield, generally 20, according to the present invention, has a stowed configuration generally adjacent craft 22 and a deployed configuration at some fixed distance 24 from craft 22. Shield 20 thus can assume any diameter from its stowed configuration to its deployed configuration. Distance or space 24 between shield 20 and craft 22 is crucial to the present invention. See, for example, exemplary nominal distance indications on present Figure 8. By deploying shield 20 some fixed distance 24, impinging particles, such as meteoroids and debris, upon contact, fragment into very small low velocity particles with a lowered (noncritical) normal to spacecraft wall energy flux which essentially dissipate in space 24.

Distinct weight advantages are gained as the result of the enhanced protection afforded by increased shield spacing 24. Generally, a craft's shield must have a certain "theoretical" thickness for impingement survivability probability requirements. The shield thickness can, however, be downsized as the shield is moved away from the craft while maintaining essentially the same survivability probability.

Deployed distance 24 is also critical to the thermal isolation characteristics of shield 20. The design of deploying shield 20 at a fixed distance 24 allows for low thermal conductivity craft/shield interfaces, fewer conduction mode heat transfer paths, and conduction paths which are long and tortuous and have many contact resistances.

Shield 20, according to the present invention, comprises a plurality of relatively flexible lattice structures 26. As embodied herein and shown for example in Figure 1, lattice structures 26 generally encircle craft 22 providing essentially 360 degree structural coverage therearound. In the preferred embodiment, only four 90 degree structures 26 are utilized, significantly increasing the weight efficiency and reliability of shield 20. Structures 26 are oriented generally along the longitudinal axis of craft 22. Preferably, structures 26 are of essentially the same length as craft 22 to be enclosed. However, this is not meant to be a limitation. More than one lattice structure 26 could be used to "cover" the length of craft 22.

Lattice structures 26 exhibit a high degree of longitudinal stiffness, but are tailored to exhibit flexibility along their arclength, thus having the ability to change curvature as they are deployed out to a larger diameter. Therefore, lattice structure 26 can be restrained at a small diameter (stowed configuration) and deployed outward to a larger diameter with the flexure allowing for curvature correction and uniform radial spacing. This principle is illustrated in Figures 3 through 6.

As embodied herein and shown for example generally in Figures 2A, 2B, and 2C, lattice structure 26 preferably comprises a lattice web 27. This web 27 provides a skeletal frame to which is affixed relatively rigid upper longitudinal members 28, relatively rigid lower longitudinal members 29, and ring members 30. Rivets 31, or any other suitable attaching means, may be utilized for attaching the members. Lower longitudinal members 29 are preferably provided along the sides and centerline of lattice structure 26, as illustrated in Figures 1 and 2C. Although shown in the Figures as being affixed to the "underside" of lattice

web 27, ring member may 30 may alternately be affixed to the top side of web 27, similar to upper longitudinal member 28.

5 Because lattice structure 26 is firmly restrained against craft 22 structural framework 21 in the stowed configuration, the flexibility characteristics do not contribute to any ascent (i.e., during lift-off) vibration problems; and once deployed in the vacuum of space, there are no "forcing functions" to drive any vibrations of lattice structure 26. The degree of flexure of lattice structure 26 is accomplished by elastically straining ring members 30 from a near-to-deployed configuration shape down to the stowed configuration shape, utilizing the strain energy of ring members 30 as the curvature correcting force. This can entail varying the moment of inertia (I) of ring members 30 along the arclength such that structure 26 deploys concentrically with respect to craft 22. Moment of inertia (I) of ring members 30 may be varied by drilling holes 33 or slots in ring member 30, as depicted in Figure 2A.

20 Shield 20 of the present invention further includes replaceable shield panels, generally 32, removably attached to lattice structure 26, forming a lattice panel 34. As embodied herein and shown for example in the Figures, shield panels 32 are generally attached to lattice web 27, essentially "filling in" lattice structure 26. Shield panels 32 may also be attached directly to upper longitudinal members 28 and ring members 30. Together, lattice structure 26 and shield panels 32 form a continuous lattice panel 34. A plurality of lattice panels 34 generally encircle craft 22 providing 360 shielding protection. In a preferred embodiment of the present invention, four 90 degree lattice panels are utilized.

30 Shield panels 32 are essentially sacrificial panels providing meteoroid and debris protection to craft 22. Panels 32 can be fabricated from many known suitable

materials, such as 6061-T6 aluminum, with weight and strength being the major design considerations. Should a panel 32 be damaged by meteoroid "hits", it is readily replaceable by on-orbit changeouts. Hence, means are provided for removably securing shield panels 32 to lattice structure 26. In the preferred embodiment of the present invention, shield panels 32 are attached to lattice structure 26 with quick acting floating fasteners 66, as depicted in Figures 3 and 5, which allow shield panels 32 to flex in conjunction with lattice structure 26. Fasteners 66 are generally known in the art and any such fastener providing a relatively bind-free interface between shield panels 32 and lattice structure 26 are within the scope of this invention.

Shield 20 of the present invention further comprises actuation means which secure lattice structure 26 relative craft framework 21 and provide means for driving lattice panels 34 outward fixed distance 24 from craft 20. As embodied herein and shown for example in Figures 2 through 8, the actuation means may preferably comprise a plurality of spring actuators 42 for each lattice panel 34. Spring actuators 42 dictate fixed distance 24 lattice panels 34 will be deployed from craft 22.

Spring actuators 42 may comprise a spring loaded pivot arm 44 which has a pivot end 46 and a shield end 48. Pivot end 46 is nonslidably attached to craft framework 21, with shield end 48 being slidably attached to lattice structure 26, preferably to lower longitudinal members 29 along the sides and centerline of lattice structure 26. A slotted trunnion bracket 54 or other equivalent means may be employed for slidably attaching shield end 48. Bracket 54, or other attaching means, may constitute an integral component of lower longitudinal member 29 or, in the alternative, comprise a separate component affixed to longitudinal member 29.

Telescoping arm 52 resides generally within a spring mechanism 50, being pivotally attached to pivot arm 44, preferably with monoball link 94 or equivalent means, and craft framework 21. Telescoping arm 52 locates spring assembly 50 between pivot arm 44 and craft framework 21, with spring assembly 50 providing the motive force urging pivot arm 44 away from craft 22. The degree of "telescoping" of arm 52 defines fixed distance 24 lattice panels 34 will be deployed. Particularly, telescoping arm 52 has stop elements at the proper extension length thereof limiting its telescoped length and defining the radial distance shield end 48 of pivot arm 44 will be deployed outward from craft 22.

Spring assembly 50 may comprise a helical compression spring 60, as depicted in Figure 8, or concentric double spring assembly (not depicted in the Figures). Any other suitable spring arrangement sufficing for urging pivot arm 44 away for craft 22 is within the scope of the invention.

In the preferred embodiment of the present invention, lattice panels 34 are thermally isolated from craft 22. This is accomplished by providing a minimum amount of thermal interfaces between panels 34 and craft framework 21. "Minimum" here refers to the least number of interface points required to adequately secure panels 34 relative the craft. As embodied herein and shown for example in Figures 7A, 7B, and 8, spring actuators 42 are affixed to craft framework 21 and lattice structure 26 with thermally isolating bearing elements 38. Bearing elements 38 provide nearly the absolute minimum thermal interface 36 between panels 34 and craft framework 21. Use of ball, roller, or needle bearings provide an infinitely small conduction path cross-sectional area at the bearing element 38 roller/race interface. Thermal/vacuum tests have recorded conductance

values as low as 0.01 btu/hr-degree F for appropriately sized deep groove ball bearings.

5 In another preferred embodiment of this invention, as depicted generally in Figure 1 and schematically in Figures 4 and 6, spring actuators 42 comprise a plurality of medial actuators 56 and a plurality of side actuators 58. This embodiment also stands alone (i.e., can be used apart from other present features) as a preferred embodiment of the present invention. Medial actuators 56 are oriented 10 longitudinally relative craft 22 and lattice panel 34, being slidably attached to lattice structure 26 at its centerline lower longitudinal member 29. Side actuators 58 are oriented perpendicular to medial actuators 56 and spaced generally equally along the side edges of lattice 15 structure 26, preferably along the side lower longitudinal members 29, and attached in the same manner as medial actuators 56, both of which are preferably attached with thermally isolating bearing elements 38.

20 It is preferred that medial actuators 56 be employed in opposing groups of two to ensure a balanced deploying force. In a most preferred embodiment of the present invention, four medial actuators 56 and eight side actuators 58 are provided for each lattice panel 34.

25 Upon actuation, or deployment of shield 20, medial actuators 56 translate the center of lattice panel 34 radially outward away from craft 20. Side actuators 58 simultaneously urge the sides of lattice panel 34 radially outward the same extent that medial actuators 56 translate the centerline of panel 34, thereby providing for uniform 30 change in curvature of lattice panel 34 to its deployed diameter. This operation is illustrated generally in Figures 3 through 8.

35 In the most preferred embodiment of the present invention, shield 20 further includes hinge assemblies 62 disposed between and affixed to lattice panels 34. As

embodied herein and shown for example in Figures 3 through 7B, hinge assembly 62 preferably comprises an accordion-like or butterfly hinge 86 which is foldable when shield 20 is in its stowed configuration and expandable between lattice panels 34 when shield 20 is deployed. By expanding between lattice panels 34, hinge 86 comprises make-up shield material 64 between lattice panels 34, thus providing for continuous 360 degree shielding around craft 20.

It should be understood that hinge assembly 62 is not limited to mechanical type hinges. A flexible element or like material may also be employed as make-up shielding 64. The only requirement is that the material fold generally upon itself when shield 20 is stowed and expand between lattice panels 34 upon deployment of shield 20.

Shield 20 of the present invention further comprises restraining means holding shield 20 relatively tight against craft 22 prior to deployment thereof. As embodied herein and shown for example in Figures 1 through 8, the restraining means preferably comprises at least two tension bands 68 which generally encircle and hold shield 20 against framework 21 of craft 22. In a most preferred embodiment, four tension bands 68 are provided. Tension bands 68 generally encircle craft 22 over ring members 30 of lattice structure 26. The tension force in bands 68 produces a radial load which, with the aid of locating surfaces (*i.e.*, between ring members 30 and craft structure 21), holds lattice panels 34 tight against suitable craft framework structural elements. The amount of tension in an individual band can be as high as 6,000 pounds. The amount of radial holding force depends on the locating surface contact areas between ring members 30 and craft structure 21 and is variable by design.

Shield restraining means may further comprise a pyrotechnically actuated trunnioned tension bolt

assembly 70, as illustrated for example in Figures 3 and 5. Bolt assembly 70 holds tension band 68 together when shield 20 is in its stowed configuration and, upon actuation thereof, separates releasing band 68 and allowing actuators 42 to deploy lattice panels 34. Tension bolt assembly 70 can be accurately verified with an electrical strain measuring device (resistance gauge) so that repeatable reliable tensioning and uniform structural loading between band 68 can be attained. The pyrotechnic devices of assembly 70 are preferably wired parallel and redundant ensuring simultaneous release of bands 68.

In the preferred embodiment of the present invention, shield 20 may further include a plurality of flexure close-out assemblies 72. The operation and apparatus of flexure close-out assemblies 72 further stand alone as a preferred embodiment of the present invention. As embodied herein and shown for example in Figures 9 through 11, flexure close-out assemblies cooperate to cover, or "close-out", annular areas 74 that exist around the ends of craft 22 between shield 20 and craft 22 when shield 20 is in its deployed configuration. In a most preferred embodiment, eight flexure close-out assemblies 72 are provided, four for each end. Flexure close-out assembly 72 preferably comprises a plurality of flexure elements 76 having cut-out reliefs 78 which permit elements 76 to "flex" with deploying shield 20 and thus track the change in curvature of lattice panels 34. Use of flexure elements 76 for covering annular areas 74 circumvents much complexity, allows for friction free deployment, and provides significant weight savings.

Flexure elements 76 are attached to lattice panels 34 at the general ends thereof, preferably to the end ring members 30 of lattice structure 26. Flexure elements 76 are attached between each of cut-out reliefs 78. At their other end, flexure elements 76 are attached to craft's 20 close-out assembly 88.

In the preferred embodiment as shown in Figure 11, bushing arrangement 80 is provided for attaching flexure elements 76 to lattice structure 26. Bushing arrangement 80 comprises thrust surface 82 and spring mechanism 84 which urges flexure element 76 against thrust surface 82 in a nonbinding interface due to adequate diametrical clearance 83 in the bushing flexure element hole.

It is to be understood by those of ordinary skill in the art that the foregoing specification and drawings discussed with reference thereto are only exemplary embodiments of the present invention, with all such language being by way of example only. Individual features and aspects of the foregoing exemplary embodiments may be varied for accommodating alternative applications, all without departing from the spirit and scope of the present invention set forth in the appended claims.

BRIEF ABSTRACT OF THE DISCLOSURE

A thermally isolated deployable shield for spacecraft is provided utilizing a plurality of lattice panels stowable generally against the craft and deployable to some fixed distance from the craft. The lattice panels are formed from replaceable shield panels affixed to lattice structures. The lattice panels generally encircle the craft providing 360 degree coverage therearound. Actuation means are provided from translating the shield radially outward from the craft and thermally isolating the shield from the craft. The lattice panels are relatively flexible, allowing the shield to deploy to variable diameters while retaining uniform curvature thereof. Restraining means are provided for holding the shield relatively tight in its stowed configuration. Close-out assemblies provide light sealing and protection of the annular spaces between the deployed shield and the crafts end structure.

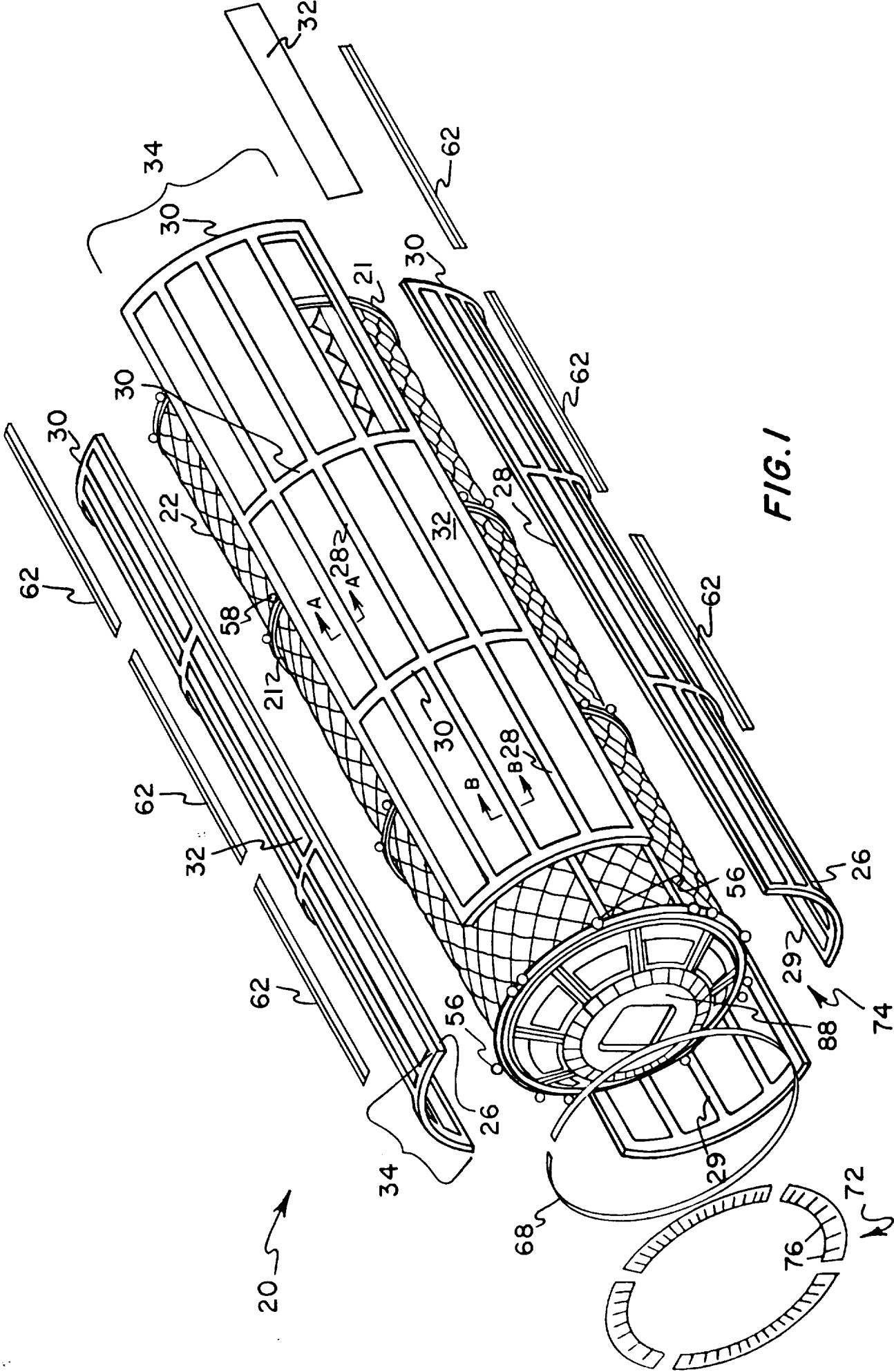


FIG. 1

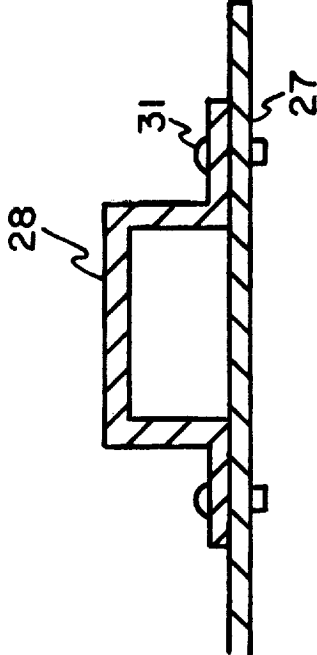


FIG. 2B A-A

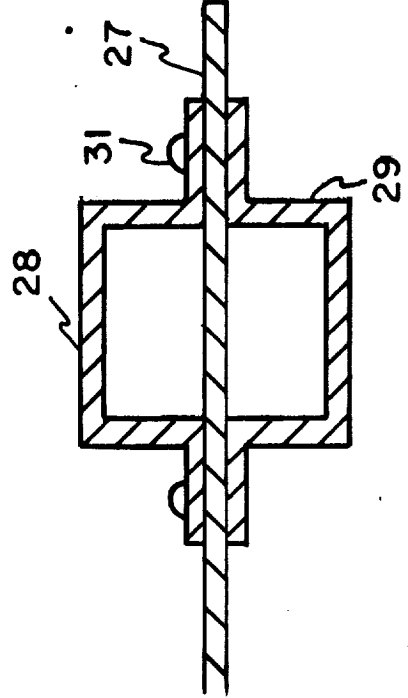


FIG. 2C B-B

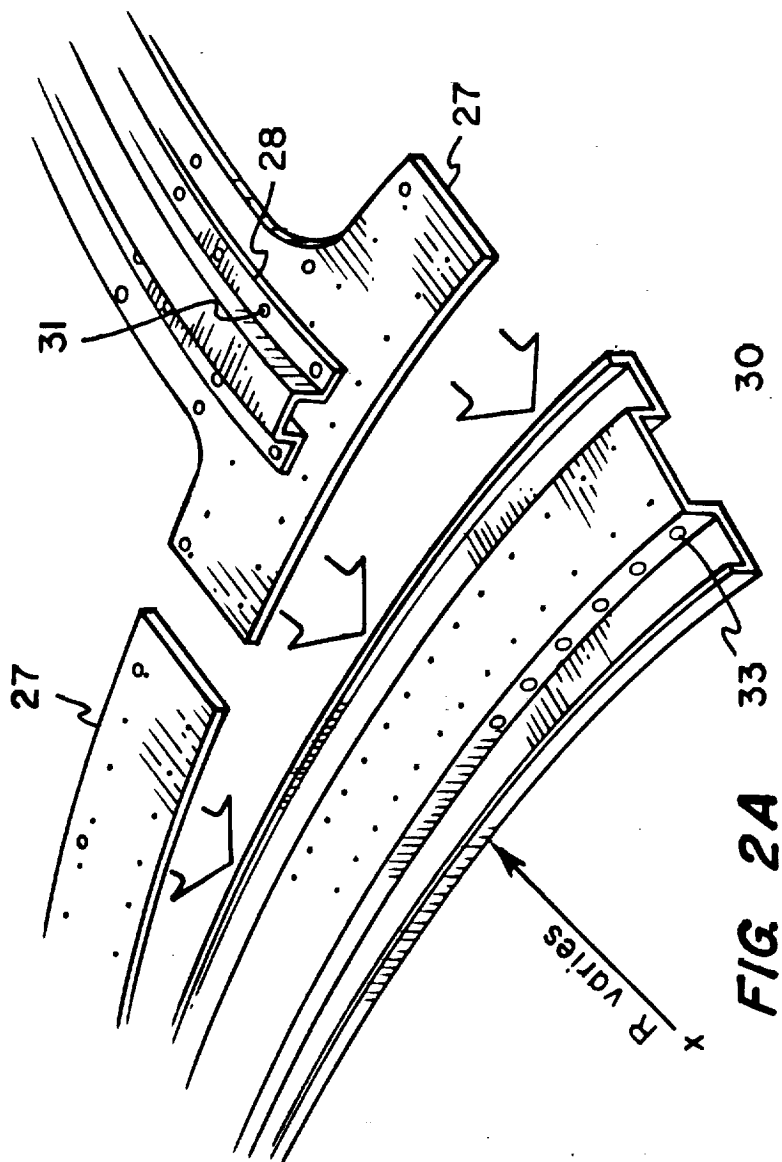


FIG. 2A

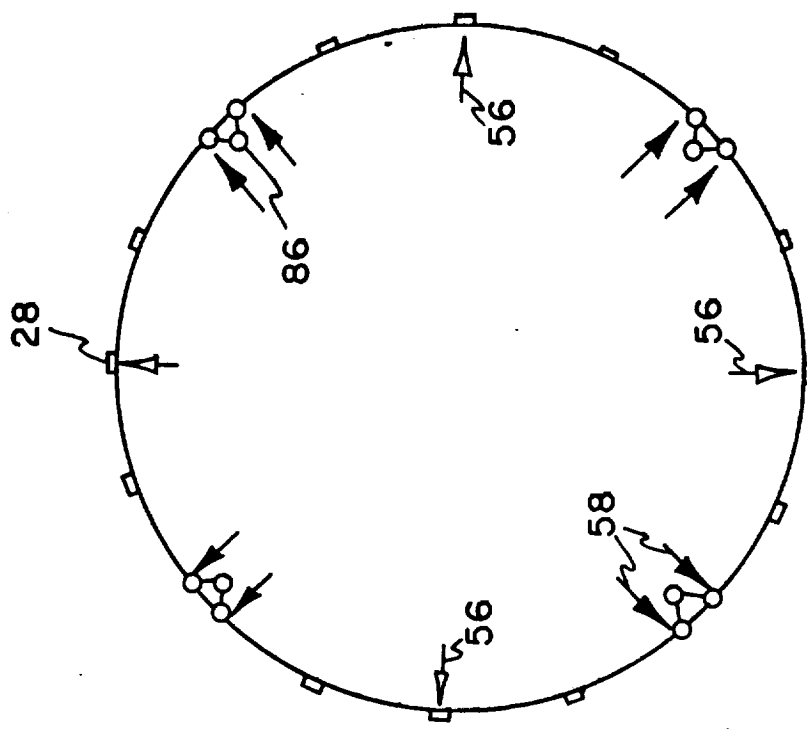


FIG. 4

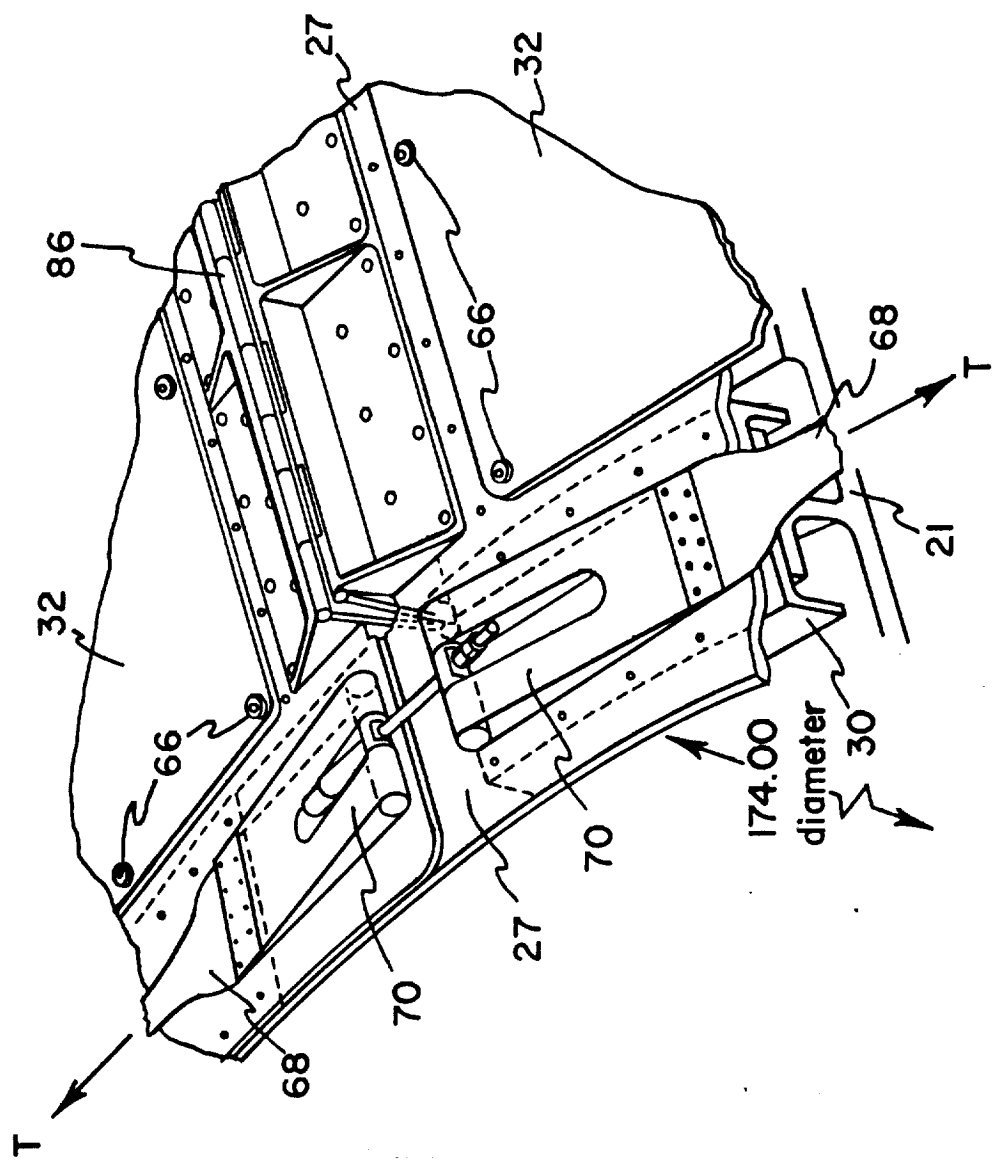


FIG. 3

FIG. 5

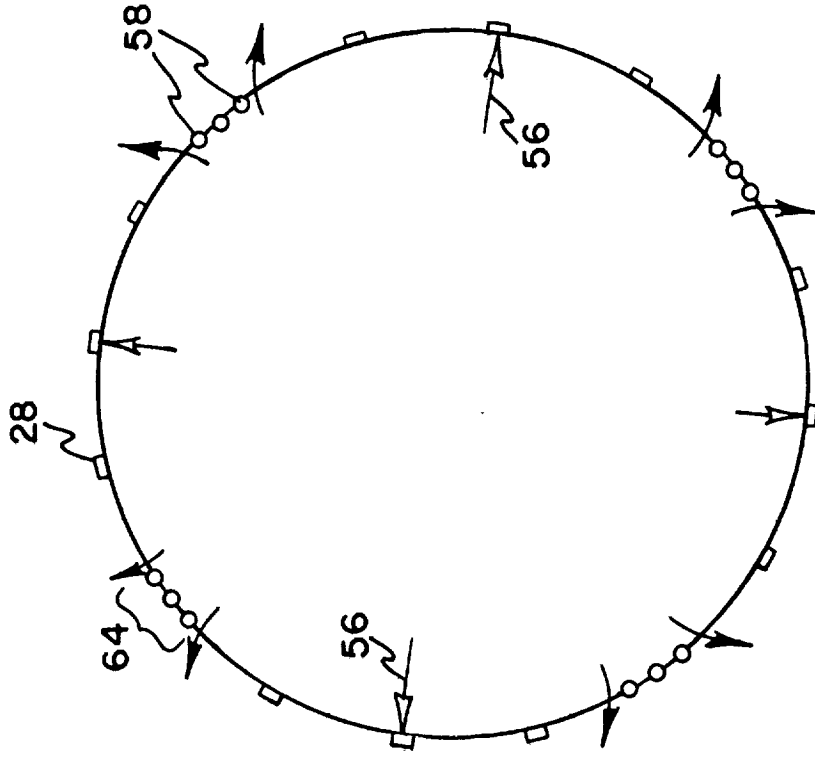
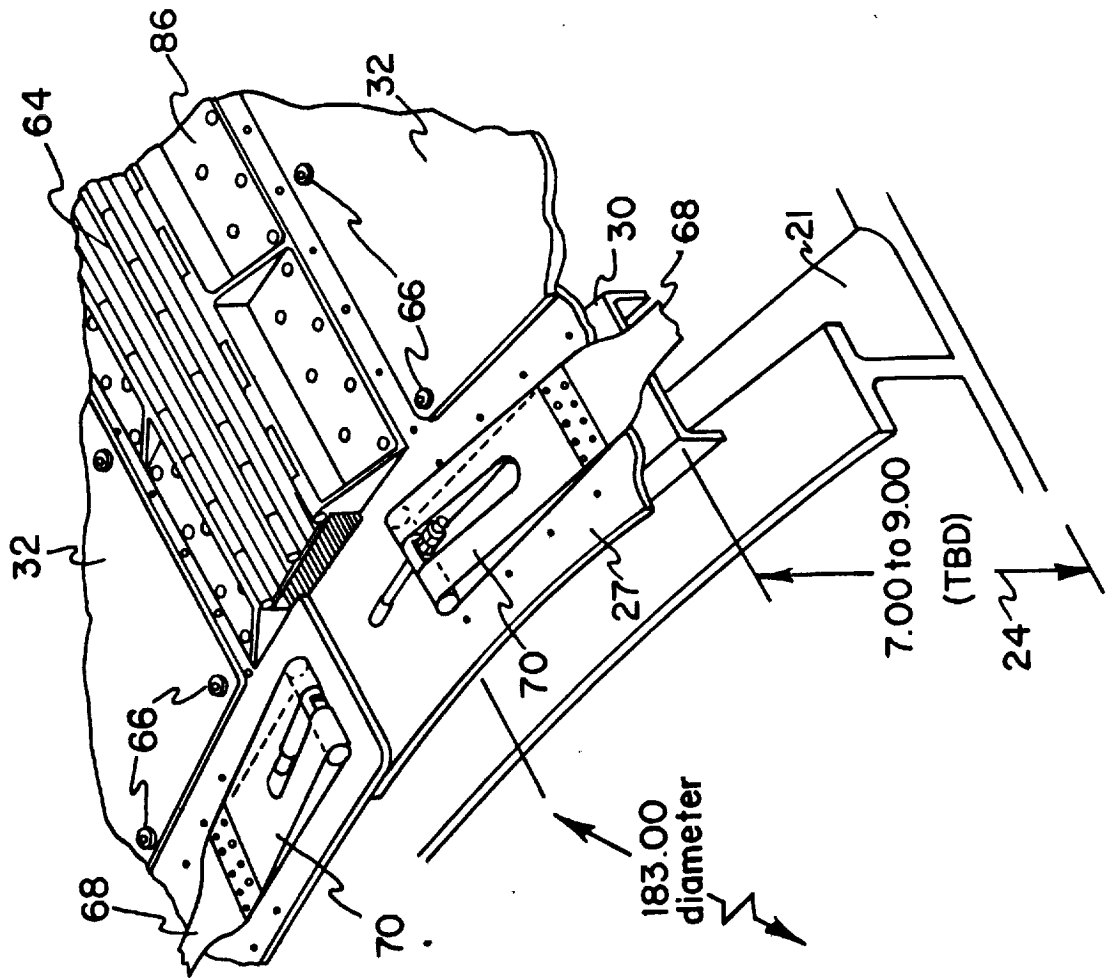
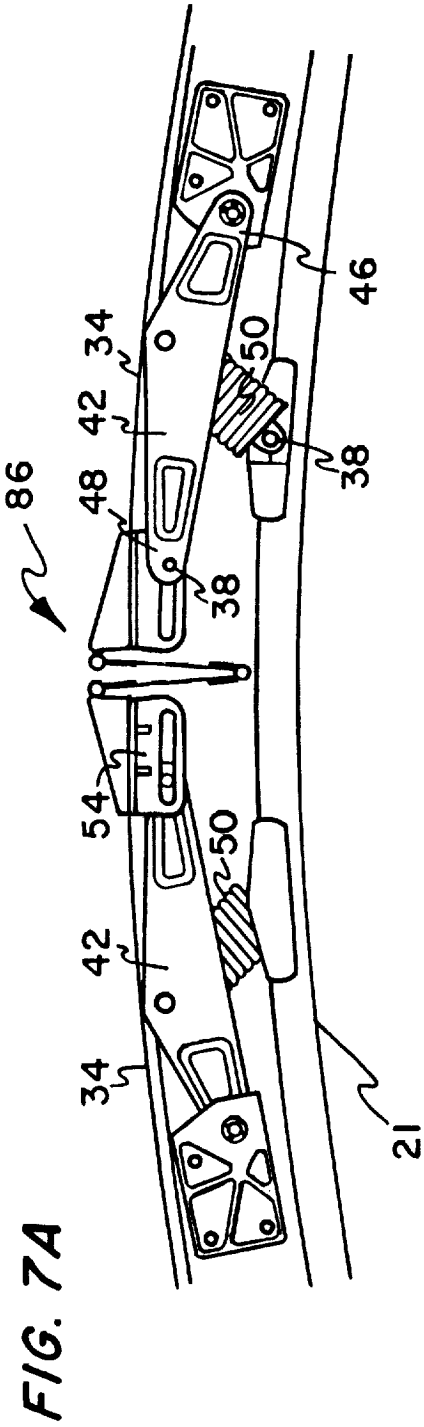
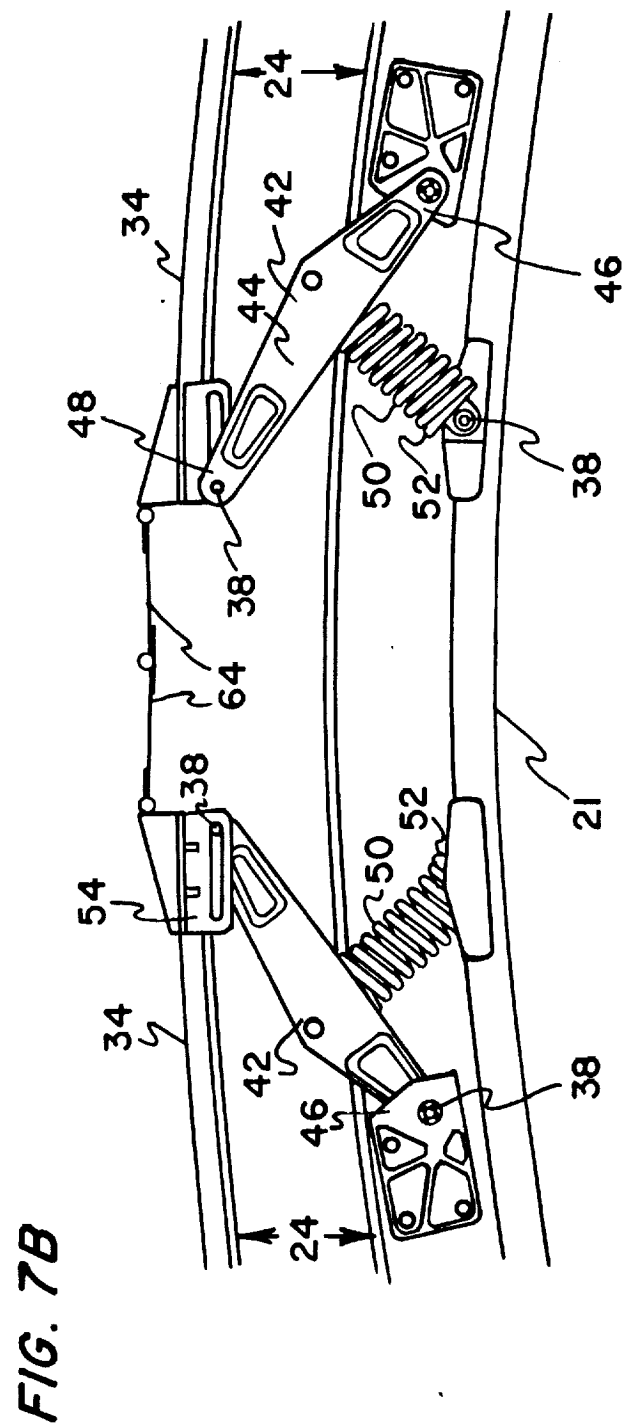
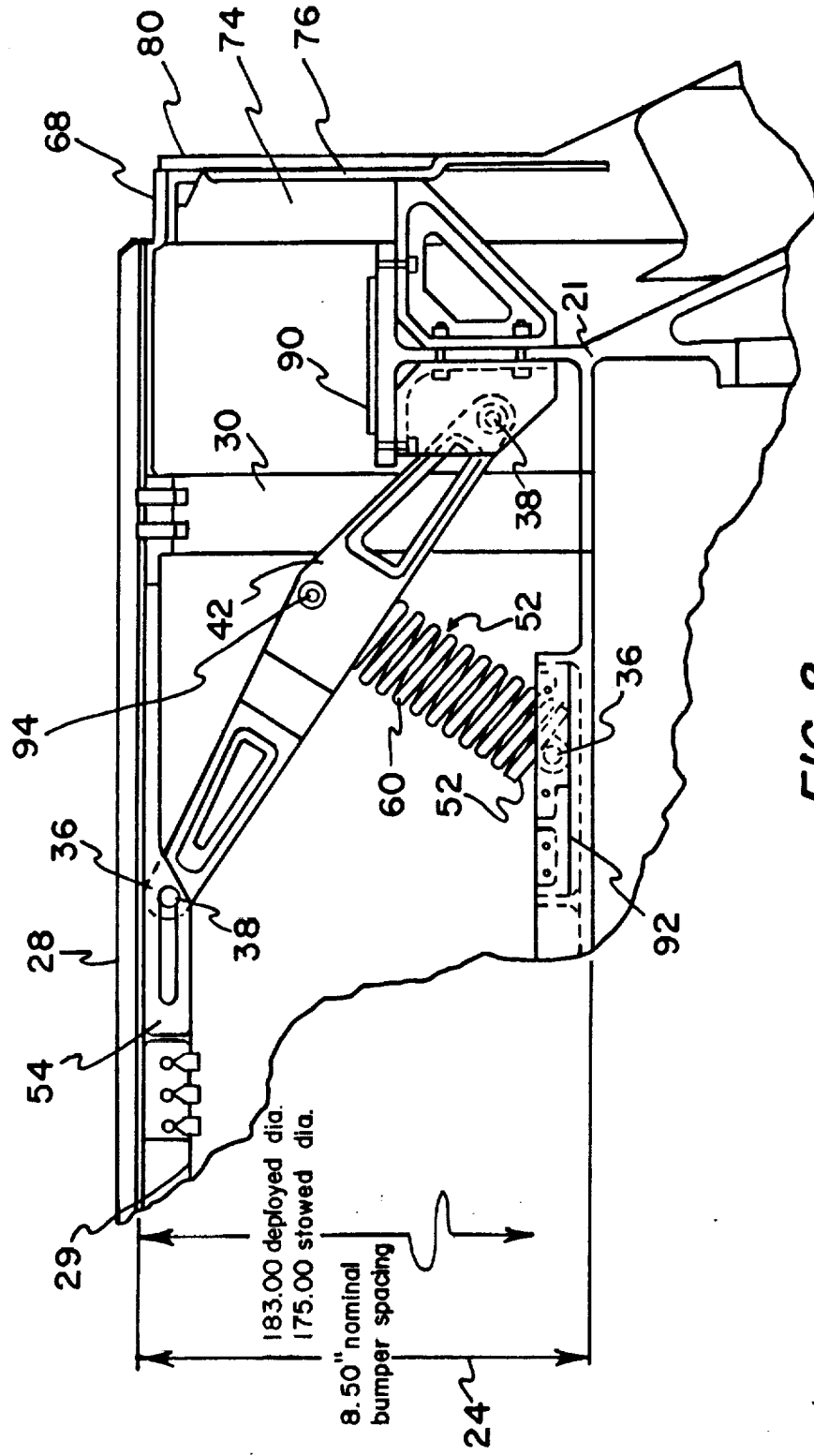


FIG. 6



STOWED





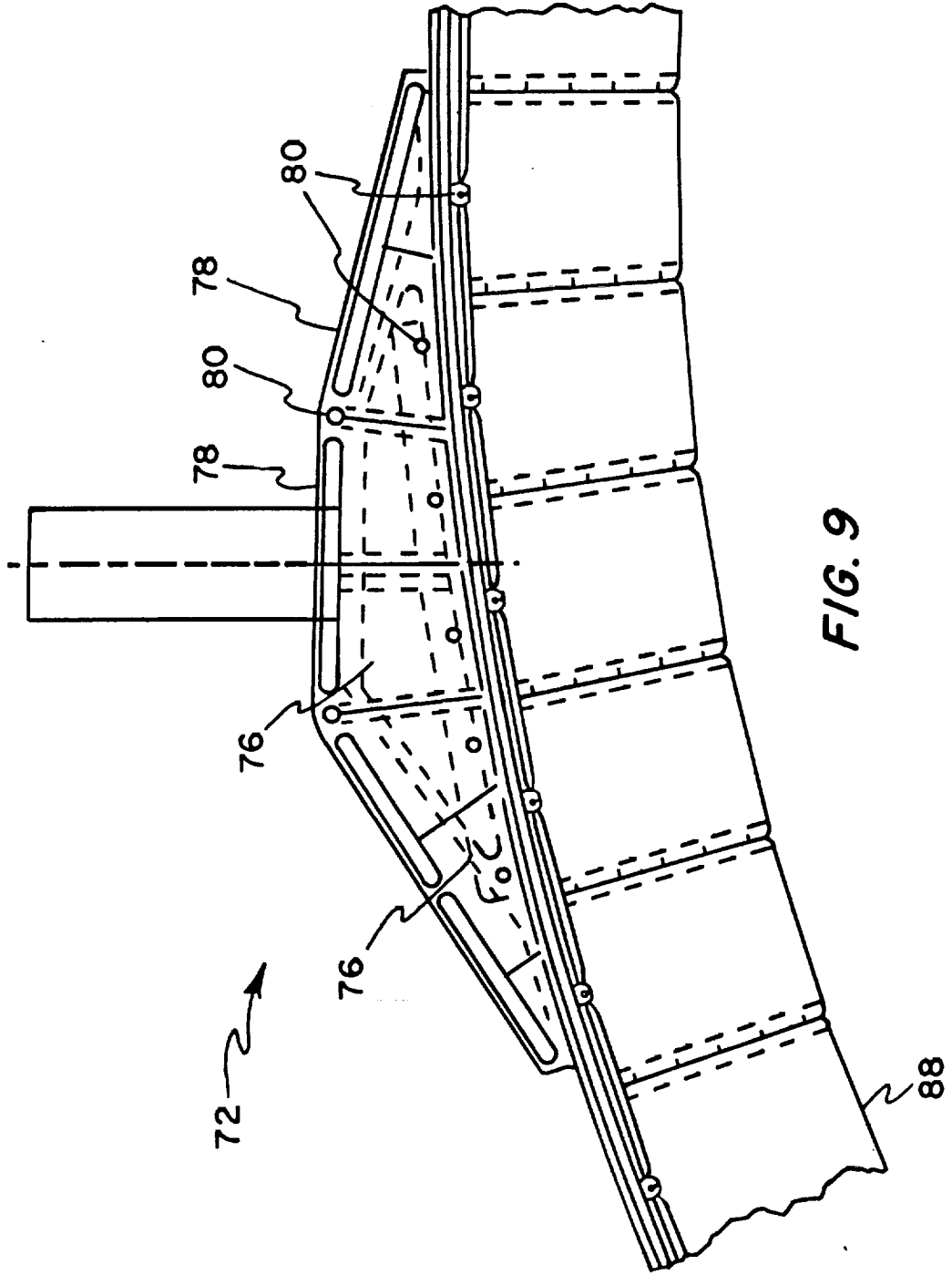


FIG. 9

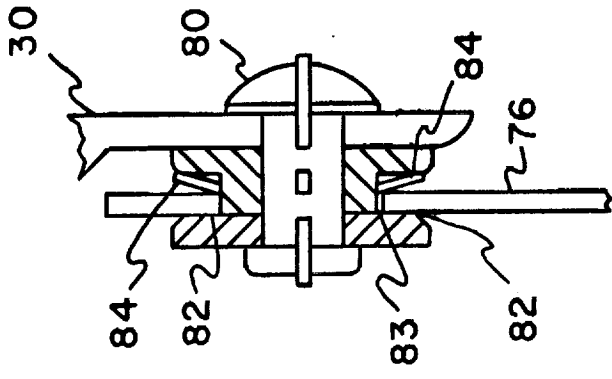


FIG. 11

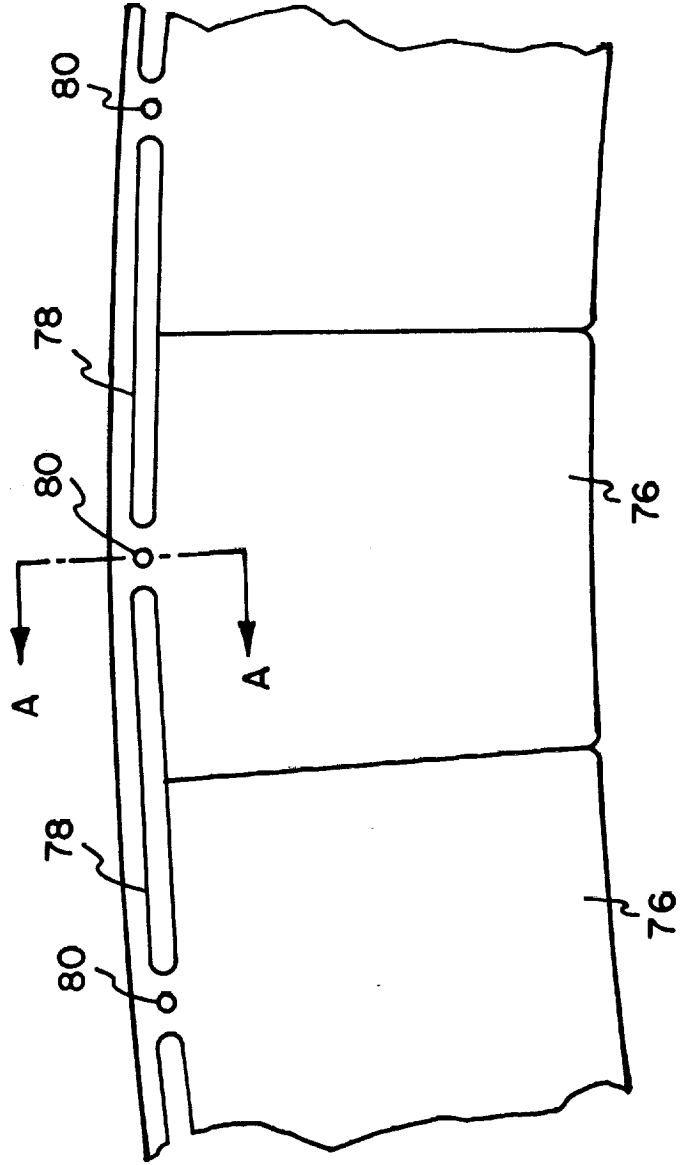


FIG. 10