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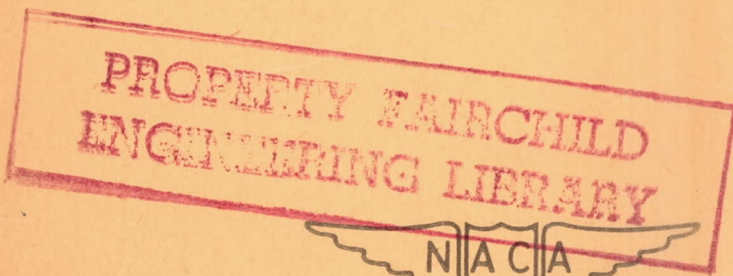
TECHNICAL NOTE

No. 1143

CHARTS FOR DETERMINING THE CHARACTERISTICS OF  
SHARP-NOSE AIRFOILS IN TWO-DIMENSIONAL  
FLOW AT SUPERSONIC SPEEDS

By H. Reese Ivey, George W. Stickle,  
and Alberta Schuettler

Langley Memorial Aeronautical Laboratory  
Langley Field, Va.



Washington  
January 1947

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CHARTS FOR DETERMINING THE CHARACTERISTICS OF  
SHARP-NOSE AIRFOILS IN TWO-DIMENSIONAL  
FLOW AT SUPERSONIC SPEEDS

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and Alberta Schuettler

September 1947

Since Technical Note No. 1143 was completed, the need for a more extensive version of table I, "Values of Local Mach Number, Pressure Ratio, and Pressure Coefficient across Shock Waves," has become apparent. This table is now available in expanded form; and a copy of the expanded table is included in this supplement to supersede the original table I.

Errors in the original publication are as follows:

Page 11: The first sentence of the last paragraph should begin "Tables I and III . . ." instead of "Tables I and II . . ."

Corrections in tables II and III are as follows:

Table II.--

$M_b$	$-\beta$	$M_a$
2.3	23°	3.4225
2.6	10°	3.0867
4.3	3°	4.5658
4.7	7°	5.4669
4.7	8°	5.5922
4.7	9°	5.7240
6.2	9°	7.9200

Table III.--

$M_b$	$-\beta$	$p_a/p_b$
1.4	14°	0.49071
4.7	24°	.02350

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Cleveland, Ohio

Figure 13.— Each vertical space along the scale label for pressure coefficient  $\Delta p/q$  should represent 0.125 instead of 20; thus, the vertical scale should appear:

-.500  
-.375  
-.250  
-.125  
0  
.125  
.250  
.375  
.500  
.625  
.750  
.875  
1.000

TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO,  
AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES

$\theta$ (deg)	$\beta$ (deg)	$M_0$	$M_1$	$\frac{P_1}{P_0}$	$\frac{\Delta p_{A,B}}{q_0}$	$\theta$ (deg)	$\beta$ (deg)	$M_0$	$M_1$	$\frac{P_1}{P_0}$	$\frac{\Delta p_{A,B}}{q_0}$
8	0	7.18546	7.18546	1.00000	0	18	0	3.23604	3.23604	1.00000	0
	1	7.80125	7.57479	1.20854	.00490		1	3.35720	3.29834	1.08900	.01128
	2	8.60430	8.07261	1.50622	.00977		2	3.49146	3.36878	1.19145	.02244
9	0	6.39264	6.39264	1.00000	0	4	3	3.64174	3.44868	1.31087	.03349
	1	6.87407	6.69357	1.18235	.00551		4	3.81150	3.53948	1.45184	.04443
	2	7.48067	7.06770	1.43092	.01100		5	4.00564	3.64318	1.62091	.05528
10	0	5.75871	5.75871	1.00000	0	6	6	4.23083	3.76221	1.82753	.06604
	1	6.14639	5.99785	1.16236	.00614		7	4.37062	3.89495	2.16071	.07937
	2	6.62119	6.28846	1.37565	.01224		8	4.51743	4.05749	2.44888	.08734
11	0	5.24082	5.24082	1.00000	0	10	9	5.21582	4.29598	2.86418	.09789
	1	5.55967	5.43448	1.14629	.00676		10	5.72912	4.47828	3.49010	.10838
	2	5.94262	5.66634	1.33336	.01349		11	6.42781	4.75694	4.43641	.11882
12	0	4.80977	4.80977	1.00000	0	12	12	7.46024	5.10668	6.03381	.12921
	1	5.07739	4.96951	1.13344	.00739		13	9.22255	5.56127	9.30925	.13956
	2	5.39281	5.25796	1.29599	.01474		19	0	3.07154	3.07154	1.00000
3	5.77302	5.58268	1.51411	.02204	1	3.18101		3.12594	1.08465	.01195	
4	6.24375	5.95433	1.79936	.03299	2	3.30163		3.18741	1.18134	.02377	
13	0	4.44543	4.44543	1.00000	0	3	3	3.43552	3.25671	1.29288	.03545
	1	4.67339	4.57891	1.12271	.00803		4	3.58559	3.33514	1.42320	.04702
	2	4.93834	4.73465	1.27307	.01600		5	3.75551	3.42421	1.57743	.05849
14	0	5.63112	5.63112	1.00000	0	6	6	3.95028	3.52573	1.76303	.06985
	1	6.10349	5.99892	1.20328	.03960		7	4.17683	3.64210	1.99073	.08113
	2	6.71567	6.42986	1.49588	.04738		8	4.44535	3.77664	2.27703	.09232
15	0	4.13360	4.13360	1.00000	0	8	9	4.77075	3.93351	2.64786	.10343
	1	4.33044	4.24248	1.11376	.00867		10	5.17739	4.11880	3.14812	.11448
	2	4.56638	4.37683	1.25087	.01726		11	5.70582	4.34078	3.89229	.12547
16	0	3.86369	3.86369	1.00000	0	12	12	6.43414	4.61223	4.95273	.13640
	1	4.03562	3.95993	1.10614	.00931		13	7.53249	4.95273	6.84971	.14729
	2	4.14555	3.98829	1.23231	.01931		14	9.48326	5.39507	10.95449	.15813
17	0	4.55655	4.19791	1.38489	.02770	20	0	2.92381	2.92381	1.00000	0
	1	4.71842	4.34601	1.57329	.03679		1	3.02334	2.97148	1.08078	.01263
	2	5.03145	4.51957	1.81179	.04587		2	3.13239	3.02492	1.17239	.02510
18	0	5.41400	4.72539	2.12408	.05479	3	3	3.25271	3.08575	1.27725	.03744
	1	5.89618	5.04264	2.55029	.06370		4	3.38650	3.15392	1.39847	.04964
	2	6.53211	5.27618	3.16796	.07258		5	3.53676	3.23092	1.54042	.06172
19	0	4.26224	3.65781	1.44339	.08143	4	6	3.70716	3.31824	1.70890	.07369
	1	4.82181	4.14628	1.91549	.09023		7	3.90296	3.41765	1.91224	.08555
	2	5.03145	4.51957	2.18179	.09587		8	4.13158	3.53157	2.16293	.09732
20	0	3.62792	3.62792	1.00000	0	9	9	4.40349	3.66307	2.47965	.10901
	1	3.77964	3.71064	1.09962	.00996		10	4.73461	3.81626	2.89260	.12061
	2	3.95037	3.80507	1.21661	.01983		11	5.15114	3.99719	3.45455	.13215
21	0	4.14479	3.91330	1.35610	.02961	12	12	5.69739	4.21372	4.26333	.14362
	1	4.36900	4.03782	1.52531	.03931		13	6.46071	4.47840	5.52986	.15503
	2	4.63202	4.18238	1.73515	.04895		14	7.63854	4.81027	7.79624	.16640
22	0	4.94653	4.35137	2.00221	.05851	15	15	9.83090	5.24117	13.02305	.17772
	1	5.33290	4.51514	2.35426	.06803		16	10.29326	5.09813	15.70811	.18831
	2	5.82325	4.79190	2.83915	.07748		17	11.78248	4.67780	18.90838	.19853
23	0	6.47624	5.08655	3.55104	.08659	18	18	14.29326	5.09813	22.90813	.20831
	1	7.40814	5.45674	4.69797	.09626		19	16.78248	4.67780	27.90838	.21853
	2	8.90088	5.93848	6.85994	.10559		20	19.78248	4.67780	34.90838	.22853
24	0	3.42032	3.42032	1.00000	0	21	0	2.66944	2.66944	1.00000	0
	1	3.55532	3.49188	1.09391	.01061		1	2.75318	2.70504	1.07435	.014012
	2	3.70612	3.57318	1.20313	.02113		2	2.84397	2.74795	1.15755	.027827
25	0	3.87615	3.66580	1.33170	.03154	3	3	2.94308	2.79453	1.25145	.041472
	1	4.07015	3.77171	1.48543	.04186		4	3.05197	2.84677	1.35832	.054956
	2	4.29455	3.89345	1.67261	.05210		5	3.17242	2.90542	1.48108	.068287
26	0	4.55860	4.03450	1.90576	.06227	4	6	3.30680	2.97131	1.62361	.081470
	1	4.87573	4.19924	2.20412	.07236		7	3.44826	3.04552	1.79138	.094530
	2	5.26736	4.39418	2.60026	.08240		8	3.60387	3.12950	1.99170	.107446
27	0	5.76803	4.62799	3.15127	.09237	5	9	3.83033	3.22498	2.23535	.12029
	1	6.44180	4.91440	3.97167	.10230		10	4.06462	3.33411	2.53821	.13301
	2	7.41936	5.27398	5.32300	.11219		11	4.34571	3.45998	2.92528	.14564
28	0	9.03211	5.73979	7.96901	.12204	6	12	4.69181	3.60650	3.43734	.15817
	1	10.21166	6.18335	10.21166	.13193		13	5.13349	3.79221	4.14788	.17065
	2	11.66166	6.66166	11.66166	.14182		14	5.72501	3.98579	5.19944	.18304
29	0	13.11166	7.17166	13.11166	.15171	7	15	6.57835	4.23809	6.91836	.19538
	1	14.56166	7.65016	14.56166	.16160		16	7.97050	4.55413	10.23118	.20765
	2	16.01166	8.12866	16.01166	.17149		17	10.91166	4.96439	19.32636	.21989

TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

$\theta$ (deg)	$\beta$ (deg)	$M_0$	$M_1$	$\frac{P_1}{P_0}$	$\frac{\Delta P_{a,b}}{q_b}$	$\theta$ (deg)	$\beta$ (deg)	$M_0$	$M_1$	$\frac{P_1}{P_0}$	$\frac{\Delta P_{a,b}}{q_b}$
23	0	2.55931	2.55931	1.00000	0	27	5	2.54085	2.32939	1.38571	.08535
	1	2.63663	2.59191	1.07155	.014703		6	2.62517	2.36445	1.45044	.10167
	2	2.72021	2.62854	1.15131	.029212		7	2.71711	2.40346	1.60857	.11776
	3	2.81097	2.66964	1.24071	.043519		8	2.81810	2.44693	1.74297	.13365
	4	2.91011	2.71559	1.34175	.057649		9	2.92984	2.49534	1.89742	.14935
	5	3.01917	2.76708	1.45693	.071610		10	3.05449	2.54944	2.07680	.16488
	6	3.14004	2.82478	1.58952	.085414		11	3.19487	2.60990	2.28776	.18023
	7	3.27510	2.88954	1.74384	.099068		12	3.35498	2.67778	2.53991	.19544
	8	3.42766	2.95435	1.92598	.11259		13	3.53985	2.75428	2.84643	.21051
	9	3.60192	3.04481	2.14416	.12628		14	3.75706	2.84104	3.22752	.22544
	10	3.80388	3.13837	2.41057	.13927		15	4.01751	2.93990	3.71441	.24025
	11	4.04200	3.24533	2.74332	.15244		16	4.33830	3.05365	4.35596	.25496
	12	4.32897	3.36862	3.17120	.16551		17	4.74742	3.18578	5.25280	.26956
	13	4.68441	3.51192	3.74184	.17850		18	5.29553	3.34120	6.57644	.28408
14	5.14181	3.68084	4.54240	.19141	19	6.08998	3.52689	8.73972	.29851		
15	5.76160	3.88284	5.74666	.20424	20	7.37622	3.75323	12.91636	.31288		
16	6.67259	4.12943	7.76366	.21702	21	10.10296	4.03644	24.37689	.32718		
17	8.21184	4.43824	11.84447	.22974							
24	0	2.45897	2.45897	1.00000	0	28	0	2.13006	2.13006	1.00000	0
	1	2.53038	2.48739	1.06915	.015428		1	2.18546	2.14729	1.06148	.01839
	2	2.60761	2.51974	1.14773	.030617		2	2.24436	2.16689	1.12856	.03646
	3	2.69120	2.55602	1.23122	.045607		3	2.30713	2.18917	1.20202	.05422
	4	2.78202	2.59660	1.32117	.060389		4	2.37434	2.21407	1.28293	.07170
	5	2.88137	2.64194	1.43578	.074985		5	2.44661	2.24198	1.37252	.08890
	6	2.99085	2.69267	1.55985	.089410		6	2.52465	2.27308	1.47228	.10585
	7	3.11233	2.74947	1.70294	.10367		7	2.60945	2.30767	1.58425	.12288
	8	3.24831	2.81309	1.86990	.11778		8	2.70205	2.34617	1.71071	.13906
	9	3.40126	2.88381	2.06753	.13183		9	2.89976	2.37065	1.85492	.15285
	10	3.57857	2.96449	2.30507	.14559		10	2.91678	2.43671	2.02095	.17143
	11	3.78354	3.05730	2.59633	.15930		11	3.04296	2.48992	2.21432	.18735
	12	4.02611	3.16209	2.96196	.17291		12	3.18538	2.54937	2.44239	.20308
	13	4.31993	3.28285	3.43525	.18642		13	3.34818	2.61606	2.71591	.21866
14	4.68616	3.42314	4.07188	.19984	14	3.53684	2.69124	3.04990	.23410		
15	5.16169	3.58857	4.97374	.21317	15	3.75932	2.77404	3.46732	.24941		
16	5.81451	3.78616	6.35862	.22643	16	4.02749	2.87344	4.00428	.26459		
17	6.79454	4.02728	8.74365	.23962	17	4.35986	2.98504	4.72109	.27966		
18	8.51947	4.32922	13.84205	.25276	18	4.78758	3.11464	5.72715	.29463		
19					19	5.36603	3.26703	7.24894	.30950		
25	0	2.36619	2.36619	1.00000	0	29	0	6.22241	3.44906	9.78928	.32429
	1	2.43311	2.39158	1.06692	.016149		21	7.67074	3.67090	14.96338	.33901
	2	2.50490	2.42023	1.14078	.032053		22	11.04280	3.94841	31.18958	.35367
	3	2.58225	2.45239	1.22268	.047707						
	4	2.66584	2.48815	1.31421	.063162		0	2.06266	2.06266	1.00000	0
	5	2.75692	2.52825	1.41711	.078398		1	2.11505	2.07774	1.06001	.01916
	6	2.85665	2.57296	1.53377	.093442		2	2.17054	2.09496	1.12523	.03797
	7	2.96666	2.62291	1.66728	.10831		3	2.22957	2.11455	1.19647	.05646
	8	3.08899	2.67881	1.82163	.12301		4	2.29236	2.13521	1.27431	.07457
	9	3.22621	2.74136	2.00218	.13755		5	2.36005	2.16131	1.36067	.09251
	10	3.38185	2.81164	2.21653	.15196		6	2.43270	2.18895	1.45616	.11011
	11	3.56060	2.89100	2.47509	.16622		7	2.51121	2.21966	1.56258	.12744
	12	3.76908	2.98103	2.79352	.18036		8	2.59664	2.25386	1.68225	.14455
	13	4.01684	3.08381	3.19547	.19438		9	2.69398	2.29151	1.82347	.16209
14	4.31849	3.20211	3.71940	.20831	10	2.79292	2.33402	1.97234	.17808		
15	4.69712	3.33960	4.43072	.22214	11	2.90717	2.38101	2.15090	.19454		
16	5.19339	3.50148	5.45345	.23588	12	3.03508	2.43337	2.35935	.21081		
17	5.88489	3.69496	7.04972	.24955	13	3.17984	2.49153	2.60608	.22690		
18	6.94816	3.93093	9.89298	.26315	14	3.34576	2.55740	2.90292	.24285		
19	8.51314	4.22633	16.38741	.27670	15	3.53866	2.63123	3.26710	.25864		
26	0	2.28118	2.28118	1.00000	0	16	3.76717	2.71487	3.72486	.27429	
	1	2.34380	2.30354	1.06492	.01688	17	4.04402	2.81017	4.31785	.28982	
	2	2.41071	2.32880	1.13624	.03349	18	4.38959	2.91970	5.11702	.30524	
	3	2.48254	2.35720	1.21505	.04985	19	4.83865	3.04685	6.25337	.32055	
	4	2.55994	2.38897	1.30255	.06595	20	5.45687	3.19634	7.99871	.33576	
	5	2.64389	2.42442	1.40051	.08185	21	6.38835	3.37485	11.02428	.35090	
	6	2.73534	2.46403	1.51078	.09752	22	8.04043	3.59235	17.56086	.36595	
	7	2.83560	2.50808	1.63601	.11300						
	8	2.94639	2.55727	1.77964	.12830	0	2.00000	2.00000	1.00000	0	
	9	3.06978	2.61226	1.94606	.14342	1	2.04970	2.01309	1.05869	.01996	
	10	3.20843	2.64370	2.14123	.15838	2	2.10220	2.02817	1.12228	.03953	
	11	3.36513	2.74282	2.37366	.17319	3	2.15786	2.04536	1.19144	.05873	
	12	3.54771	2.80717	2.65513	.18786	4	2.21712	2.06489	1.26706	.07743	
	13	3.76027	2.90907	3.00338	.20241	5	2.28038	2.08676	1.35004	.09616	
14	4.01396	3.00988	3.44555	.21684	6	2.34824	2.11124	1.44166	.11442		
15	4.32457	3.12584	4.02623	.23116	7	2.42132	2.13897	1.54331	.13239		
16	4.71747	3.26064	4.82275	.24539	8	2.50046	2.16931	1.65693	.15010		
17	5.23773	3.41920	5.98393	.25953	9	2.58664	2.20269	1.78480	.16757		
18	5.97457	3.60871	7.83617	.27359	10	2.68100	2.24016	1.92978	.18479		
19	7.13932	3.83975	11.26064	.28758	11	2.78510	2.28175	2.09572	.20180		
20	9.42386	4.12892	19.74410	.30152	12	2.90088	2.32799	2.28775	.21861		
27	0	2.20269	2.20269	1.00000	0	13	3.03082	2.37953	2.51256	.23523	
	1	2.26150	2.22239	1.06314	.01764	14	3.17820	2.43701	2.77947	.25167	
	2	2.32415	2.24468	1.13222	.03497	15	3.34766	2.50149	3.10201	.26795	
	3	2.39117	2.26979	1.20820	.05202	16	3.54542	2.57407	3.49958	.28408	
	4	2.46318	2.29795	1.29226	.06881	17	3.78065	2.65620	4.00222	.30006	
					18	4.06734	2.74978	4.65845	.31592		
					19	4.42795	2.85731	5.55196	.33166		

TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND  
PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$	$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$
30	20	4.90159	3.00520	6.84081	0.34730	34	0	1.78829	1.78829	1.00000	0
	21	5.56424	3.12879	8.86354	.36284		1	1.82960	1.79493	1.05453	.023271
	22	6.58991	3.30392	12.49954	.37829		2	1.87291	1.80309	1.11302	.046028
	23	8.51233	3.51725	20.96744	.39367		3	1.91837	1.81285	1.17591	.060285
							4	1.96630	1.82427	1.24382	.090089
31	0	1.94160	1.94160	1.00000	0	35	0	1.74345	1.74345	1.00000	0
	1	1.98886	1.95286	1.05748	.02076		1	1.78316	1.74876	1.05375	.02415
	2	2.03868	1.96598	1.11958	.04110		2	1.82468	1.75553	1.11126	.04774
	3	2.09138	1.98107	1.18694	.06106		3	1.86822	1.76380	1.17296	.07079
	4	2.14730	1.99823	1.26029	.08064		4	1.91398	1.77361	1.23939	.09336
	5	2.20684	2.01760	1.34052	.09989		5	1.96221	1.78504	1.31116	.11545
	6	2.27049	2.03933	1.42871	.11880		6	2.01321	1.79815	1.38897	.13710
	7	2.33880	2.06359	1.52616	.13741		7	2.06730	1.81304	1.47368	.15834
	8	2.41247	2.09061	1.63449	.15574		8	2.12487	1.82981	1.56352	.17918
	9	2.49232	2.12062	1.75569	.17379		9	2.18638	1.84899	1.66010	.19966
	10	2.57936	2.15392	1.89230	.19160		10	2.25232	1.86953	1.78054	.21979
	11	2.67484	2.19086	2.04755	.20916		11	2.32352	1.89278	1.90548	.23960
	12	2.78034	2.23185	2.22567	.22651		12	2.40058	1.91855	2.04521	.25910
	13	2.89789	2.27739	2.43223	.24364		13	2.48453	1.94707	2.20263	.27832
14	3.03011	2.32809	2.67480	.26058	14	2.57658	1.97862	2.38143	.29727		
15	3.18052	2.38468	2.96390	.27735	15	2.67822	2.01351	2.58644	.31596		
16	3.35393	2.44806	3.31456	.29394	16	2.79139	2.05215	2.82401	.33442		
17	3.55709	2.51939	3.74909	.31038	17	2.91858	2.09498	3.10277	.35266		
18	3.79995	2.60008	4.30204	.32668	18	3.06314	2.14258	3.43466	.37069		
19	4.09778	2.69201	5.02997	.34285	19	3.22962	2.19562	3.83675	.38853		
20	4.47555	2.79760	6.03288	.35890	20	3.42442	2.25495	4.33427	.40619		
21	4.97767	2.92012	7.50125	.37484	21	3.65692	2.32162	4.96621	.42369		
22	5.63902	3.06409	9.86355	.39068	22	3.94150	2.39697	5.79616	.44103		
23	6.83565	3.23596	14.29389	.40644	23	4.30153	2.48272	6.93526	.45824		
24	9.13076	3.44528	25.63452	.42212	24	4.77837	2.58113	8.59705	.47532		
32	0	1.88707	1.88707	1.00000	0	36	0	1.70130	1.70130	1.00000	0
	1	1.93213	1.89664	1.05639	.02158		1	1.73955	1.70539	1.05304	.025040
	2	1.97956	1.90799	1.11716	.04271		2	1.77947	1.71085	1.10969	.049487
	3	2.02958	1.92112	1.18287	.06342		3	1.82126	1.71772	1.17033	.073358
	4	2.08253	1.93620	1.25420	.08373		4	1.86536	1.72606	1.23585	.096830
	5	2.13875	1.95330	1.33195	.10367		5	1.91119	1.73654	1.30563	.11953
	6	2.19868	1.97256	1.41711	.12326		6	1.95980	1.74734	1.38147	.14189
	7	2.26277	2.00680	1.51078	.14251		7	2.01124	1.76042	1.46381	.16380
	8	2.33163	2.04807	1.61445	.16146		8	2.06585	1.77522	1.55356	.18530
	9	2.40596	2.09478	1.72982	.18011		9	2.12402	1.79185	1.65178	.20639
	10	2.48665	2.07445	1.85915	.19849		10	2.18626	1.81048	1.75992	.22713
	11	2.57474	2.10730	2.00522	.21662		11	2.25305	1.83118	1.87943	.24743
	12	2.67149	2.14373	2.17150	.23450		12	2.32517	1.85438	2.01251	.26754
	13	2.77856	2.18409	2.36269	.25215		13	2.40336	1.87960	2.16155	.28728
14	2.89612	2.22897	2.58506	.26960	14	2.48870	1.90778	2.32982	.30673		
15	3.03292	2.27884	2.84700	.28684	15	2.58245	1.93889	2.52146	.32591		
16	3.18672	2.33454	3.16033	.30390	16	2.68612	1.97330	2.74161	.34483		
17	3.36459	2.39688	3.54210	.32080	17	2.80180	2.01136	2.99750	.36351		
18	3.57385	2.46700	4.01779	.33753	18	2.93216	2.05355	3.29879	.38197		
19	3.82931	2.54630	4.62734	.35413	19	3.08076	2.10040	3.65894	.40022		
20	4.13577	2.63661	5.43708	.37058	20	3.25251	2.15260	4.09738	.41827		
21	4.53321	2.74033	6.56985	.38692	21	3.45439	2.21098	4.64313	.43615		
22	5.06850	2.88280	8.24971	.40315	22	3.69669	2.27656	5.31455	.45386		
23	5.84704	3.00200	11.03386	.41927	23	3.99543	2.35065	6.26779	.47141		
24	7.13785	3.17071	16.52504	.43531	24	4.37714	2.43496	7.55977	.48883		
25	9.97414	3.37616	32.42577	.45127	25	4.89989	2.53169	9.47123	.50612		
					26	5.63298	2.66431	12.62122	.52329		
33	0	1.83608	1.83608	1.00000	0	37	0	1.70130	1.70130	1.00000	0
	1	1.87918	1.84413	1.05541	.02242		1	1.73955	1.70539	1.05304	.025040
	2	1.92442	1.85381	1.11497	.04435		2	1.77947	1.71085	1.10969	.049487
	3	1.97206	1.86350	1.17920	.06583		3	1.82126	1.71772	1.17033	.073358
	4	2.02235	1.87836	1.24872	.08688		4	1.86536	1.72606	1.23585	.096830
	5	2.07561	1.89338	1.32426	.10752		5	1.91119	1.73654	1.30563	.11953
	6	2.13221	1.91039	1.40668	.12779		6	1.95980	1.74734	1.38147	.14189
	7	2.19258	1.92947	1.49703	.14770		7	2.01124	1.76042	1.46381	.16380
	8	2.25722	1.95082	1.59657	.16727		8	2.06585	1.77522	1.55356	.18530
	9	2.32674	1.97459	1.70686	.18653		9	2.12402	1.79185	1.65178	.20639
	10	2.40188	2.00099	1.82982	.20796		10	2.18626	1.81048	1.75992	.22713
	11	2.48351	2.03026	1.96784	.22217		11	2.25305	1.83118	1.87943	.24743
	12	2.57274	2.06268	2.12398	.24259		12	2.32517	1.85438	2.01251	.26754
	13	2.67094	2.09859	2.30217	.26076		13	2.40336	1.87960	2.16155	.28728
14	2.77983	2.13838	2.50757	.27870	14	2.48870	1.90778	2.32982	.30673		
15	2.90165	2.18255	2.74709	.29643	15	2.58245	1.93889	2.52146	.32591		
16	3.03936	2.23168	3.03020	.31396	16	2.68612	1.97330	2.74161	.34483		
17	3.19689	2.28644	3.37020	.33131	17	2.80180	2.01136	2.99750	.36351		
18	3.37977	2.34775	3.78644	.34848	18	2.93216	2.05355	3.29879	.38197		
19	3.59589	2.41669	4.30816	.36549	19	3.08076	2.10040	3.65894	.40022		
20	3.85704	2.49464	4.98175	.38235	20	3.25251	2.15260	4.09738	.41827		
21	4.18188	2.58339	5.88545	.39908	21	3.45439	2.21098	4.64313	.43615		
22	4.60196	2.68528	7.16241	.41569	22	3.69669	2.27656	5.31455	.45386		
23	5.17619	2.80347	9.10556	.43218	23	3.99543	2.35065	6.26779	.47141		
24	6.03149	2.94229	12.42298	.44857	24	4.37714	2.43496	7.55977	.48883		
25	7.51479	3.10795	19.37661	.46487	25	4.89989	2.53169	9.47123	.50612		
26	11.19767	3.53991	43.22636	.48109	26	5.63298	2.66431	12.62122	.52329		

TABLE I - VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND  
PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$	$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$
36	27	6.85490	2.77549	18.77364	0.54035	39	17	2.51649	1.79765	2.75938	0.39689
	28	9.49197	2.93252	36.14927	.55732		18	2.61643	1.82749	2.99641	.41661
37	0	1.66164	1.66164	1.00000	0	20	19	2.72771	1.86043	3.27119	.43607
	1	1.69857	1.66458	1.05244	.02597		20	2.85280	1.89684	3.59373	.45529
2	1.73706	1.66883	1.10831	.05128	21	21	2.99499	1.93714	3.97793	.47427	
	3	1.77726	1.67443	1.16800		.07598	22	3.15879	1.98195	4.44366	.49304
3	1.81934	1.68140	1.23196	.10011	23	23	3.35052	2.03163	5.02033	.51161	
	4	1.86351	1.68980	1.30070		.12370	24	3.68530	2.09759	6.03867	.52999
5	1.91000	1.69967	1.37483	.14678	25	25	3.85991	2.14968	6.71741	.54821	
	6	1.95907	1.71107	1.45505		.16938	26	4.21505	2.22019	8.04246	.56627
7	2.01103	1.72408	1.54221	.19153	27	27	4.68611	2.30037	9.97979	.58418	
	8	2.06623	1.73879	1.63730		.21225	28	5.35512	2.39231	13.08374	.60196
10	2.12508	1.75530	1.74154	.23458	29	29	6.41824	2.49885	18.86698	.61961	
	11	2.18809	1.77373	1.85637		.25552	30	8.52952	2.62389	33.44886	.63716
12	2.25584	1.79422	1.98358	.27612	40	0	1.55572	1.55572	1.00000	0	
	13	2.32903	1.81694	2.12538		.29638	1	1.58940	1.55556	1.05106	.02887
14	2.40896	1.84205	2.28941	.31643	2	2	1.62431	1.55660	1.10515	.05693	
	15	2.49541	1.86986	2.46456		.33599	3	1.66120	1.55877	1.16355	.08467
16	2.59099	1.90054	2.66997	.35537	4	4	1.69841	1.56224	1.22384	.11085	
	17	2.69694	1.93445	2.90670		.37449	5	1.73791	1.56694	1.28916	.13677
18	2.81543	1.97195	3.18269	.39337	6	6	1.77912	1.57276	1.35914	.16209	
	19	2.94931	2.01350	3.50881		.41203	7	1.82243	1.57923	1.43431	.18681
20	3.10242	2.05963	3.90033	.43043	8	8	1.86797	1.58841	1.51532	.21098	
	21	3.28008	2.11101	4.37946		.44873	9	1.91599	1.59826	1.60291	.23462
22	3.48991	2.16844	4.97971	.46679	10	10	1.96679	1.60950	1.69799	.25777	
	23	3.74331	2.23295	5.75420		.48469	11	2.02072	1.62230	1.80165	.28046
24	4.05825	2.30583	6.72940	.50244	12	12	2.07818	1.63666	1.91519	.30272	
	25	4.46512	2.38873	8.25775		.52004	13	2.13960	1.65272	2.04007	.32456
26	5.02050	2.48383	10.48377	.53751	14	14	2.20564	1.67062	2.17838	.34603	
	27	5.84590	2.59406	14.27360		.55487	15	2.27684	1.69040	2.33224	.36723
28	7.27143	2.72345	22.17482	.57211	16	16	2.35412	1.71233	2.50474	.38789	
	29	10.76973	2.87776	48.84307		.58927	17	2.43844	1.73651	2.69953	.40833
38	0	1.62427	1.62427	1.00000	0	18	18	2.53106	1.76318	2.92139	.42846
	1	1.66000	1.62610	1.05191	.02691		19	2.63355	1.79260	3.17655	.44832
2	1.69720	1.68925	1.10711	.05312	20	20	2.74794	1.82507	3.47330	.46791	
	3	1.73595	1.63363	1.16595		.07867	21	2.87691	1.86290	3.82297	.48766
3	1.77646	1.63934	1.22888	.10361	22	22	3.02399	1.90062	4.24135	.50637	
	4	1.81891	1.64639	1.29635		.12796	23	3.19412	1.94465	4.75127	.52527
5	1.86348	1.65484	1.36892	.15177	24	24	3.39425	1.99364	5.38688	.54396	
	6	1.91042	1.66471	1.44727		.17507	25	3.63476	2.04837	6.20177	.56247
7	1.95999	1.67607	1.53211	.19788	26	26	3.93174	2.10980	7.28497	.58081	
	8	2.01254	1.68903	1.62441		.22023	27	4.31206	2.17916	8.79667	.59899
10	2.06839	1.70361	1.72524	.24217	28	28	4.82461	2.25802	11.05366	.61702	
	11	2.12800	1.71998	1.83585		.26369	29	5.57095	2.34844	14.79363	.63492
12	2.19194	1.73827	1.95797	.28484	30	30	6.81144	2.47200	22.19783	.65270	
	13	2.26069	1.75849	2.09337		.30562	31	9.55562	2.57613	43.84821	.67037
14	2.33512	1.77819	2.24461	.32607	41	0	1.52425	1.52425	1.00000	0	
	15	2.41604	1.80577	2.41463		.34621	1	1.55704	1.82319	1.05073	.02989
16	2.50464	1.83315	2.60744	.36605	2	2	1.59100	1.82320	1.10441	.05892	
	17	2.60229	1.86344	2.82795		.38566	3	1.62623	1.82450	1.16133	.08714
18	2.71076	1.89686	3.08280	.40492	4	4	1.66286	1.82688	1.22183	.11461	
	19	2.83237	1.93381	3.38091		.42398	5	1.70103	1.83043	1.28631	.14135
20	2.97018	1.97473	3.73452	.44281	6	6	1.74090	1.83516	1.35521	.16743	
	21	3.12834	2.02015	4.16184		.46143	7	1.78264	1.84110	1.42906	.19288
22	3.31262	2.07072	4.68594	.47985	8	8	1.82644	1.84829	1.50845	.21774	
	23	3.51443	2.12723	5.34815		.49809	9	1.87255	1.85677	1.59408	.24204
24	3.79744	2.19070	6.21029	.51616	10	10	1.92121	1.86658	1.68679	.26581	
	25	4.13102	2.26238	7.37982		.53407	11	1.97274	1.87780	1.78755	.28910
26	4.58731	2.34390	9.05803	.55184	12	12	2.02749	1.89049	1.89753	.31191	
	27	5.17392	2.43741	11.67110		.56947	13	2.08588	1.90474	2.01813	.33429
28	6.10326	2.56549	16.30565	.58699	14	14	2.14839	1.92004	2.15104	.35626	
	29	7.80868	2.67296	26.79741		.60440	15	2.21562	1.93831	2.29837	.37784
30	12.89714	2.82462	73.38925	.62171	16	16	2.28827	1.95788	2.46267	.39906	
	0	1.58902	1.58902	1.00000		0	17	2.36721	1.97951	2.64721	.41993
1	1.62367	1.58985	1.05145	.02788	18	18	2.45349	1.70337	2.85608	.44048	
	2	1.65967	1.59190	1.10605		.05500	19	2.54845	1.72968	3.09460	.46073
3	1.69713	1.59517	1.16416	.08142	20	20	2.65378	1.75868	3.36974	.48070	
	4	1.73622	1.59970	1.22617		.10718	21	2.77164	1.79068	3.69085	.50040
5	1.77709	1.60550	1.29251	.13232	22	22	2.90492	1.82601	4.07076	.51985	
	6	1.81992	1.61261	1.36371		.15687	23	3.05749	1.86509	4.52755	.53907
7	1.86495	1.62107	1.44036	.18087	24	24	3.23473	1.90844	5.08756	.55807	
	8	1.91239	1.63094	1.52317		.20436	25	3.44438	1.95667	5.79072	.57687
9	1.96255	1.64227	1.61298	.22735	26	26	3.69807	2.01053	6.70059	.59549	
	10	2.01575	1.65514	1.71076		.24989	27	4.01421	2.07098	7.92491	.61355
11	2.07236	1.66963	1.81770	.27200	28	28	4.42429	2.13921	9.66250	.63221	
	12	2.13285	1.68584	1.93523		.29370	29	4.98692	2.21677	12.32147	.65034
13	2.19774	1.70390	2.06508	.31501	30	30	5.83108	2.30570	16.90715	.66834	
	14	2.26769	1.72394	2.20939		.33597	31	7.31588	2.40871	26.70940	.68622
15	2.34345	1.74612	2.37081	.35699	32	32	11.16425	2.52958	62.42147	.70398	
	16	2.42599	1.77061	2.55271		.37689					

TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$	$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$	
42	0	1.49448	1.49448	1.00000	0	44	15	2.05902	1.50014	2.22011	0.41113	
	1	1.52646	1.49254	1.05047	.03094		16	2.12085	1.51398	2.36559	.43372	
	2	1.55953	1.49172	1.10379	.06096		17	2.18729	1.52938	2.52675	.45889	
	3	1.59084	1.48920	1.16026	.09046		18	2.25905	1.54644	2.70637	.47767	
	4	1.63013	1.49331	1.22140	.11902		19	2.33696	1.56530	2.90795	.49908	
	5	1.66642	1.49592	1.28389	.14604		20	2.42205	1.58610	3.13593	.52014	
	6	1.70602	1.50047	1.35186	.17270		21	2.51562	1.60901	3.39602	.54088	
	7	1.74533	1.50437	1.42453	.19909		22	2.61929	1.63424	3.69573	.56132	
	8	1.78759	1.51037	1.50251	.22465		23	2.73518	1.66202	4.04507	.58147	
	9	1.83196	1.51758	1.58642	.24962		24	2.86606	1.69262	4.45777	.60135	
	10	1.87871	1.52603	1.67703	.27402		25	3.01565	1.72639	4.95312	.62098	
	11	1.92809	1.53583	1.77522	.29790		26	3.18913	1.76371	5.55910	.64038	
	12	1.98043	1.54697	1.88207	.32128		27	3.39390	1.80508	6.31798	.65956	
	13	2.03608	1.55958	1.99882	.34419		28	3.64100	1.85107	7.29665	.67853	
	14	2.09550	1.57370	2.12706	.36667		29	3.94787	1.90240	8.60767	.69731	
	15	2.15919	1.58941	2.26861	.38873		30	4.34390	1.95998	10.45639	.71592	
	16	2.22779	1.60689	2.42582	.41041		31	4.88378	2.02495	13.26101	.73437	
	17	2.30202	1.62622	2.60146	.43172		32	5.60455	2.09877	16.02541	.75267	
	18	2.38280	1.64758	2.79914	.45268		33	7.06275	2.18337	27.91591	.77084	
	19	2.47126	1.67112	3.02345	.47332		34	10.41342	2.29901	60.88204	.78888	
	20	2.56883	1.69707	3.28032	.49366		45	0	1.41421	1.41421	1.00000	0
	21	2.67731	1.72566	3.57758	.51371			1	1.44426	1.40982	1.05010	.03431
	22	2.79904	1.75719	3.92581	.53350			2	1.47521	1.40648	1.10280	.06748
	23	2.93714	1.79199	4.33962	.55303			3	1.50714	1.40419	1.15836	.09960
	24	3.09586	1.83048	4.83980	.57233			4	1.54015	1.40293	1.21704	.13071
	25	3.28113	1.87316	5.45696	.59142			5	1.57434	1.40269	1.27916	.16090
	26	3.50161	1.92063	6.23811	.61030			6	1.60983	1.40348	1.34507	.19022
	27	3.77045	1.97364	7.25935	.62899			7	1.64673	1.40529	1.41516	.21871
	28	4.10893	2.03311	8.65247	.64751			8	1.68518	1.40814	1.48991	.24645
	29	4.55417	2.10024	10.66728	.66587			9	1.72536	1.41205	1.56983	.27346
	30	5.17863	2.17653	13.84204	.68408			10	1.76742	1.41703	1.65554	.29979
	31	6.14967	2.26399	19.58813	.70216			11	1.81159	1.42311	1.74775	.32549
	32	7.98864	2.38362	33.16941	.72011			12	1.85809	1.43032	1.84730	.35059
33	14.19485	2.48414	105.08535	.73795	13	1.90720		1.43872	1.95535	.37513		
14	1.95922	1.44834	2.07248	.39914	14	1.95922		1.44834	2.07248	.39914		
43	0	1.46628	1.46628	1.00000	0	46	0	1.39016	1.39016	1.00000	0	
	1	1.49755	1.46349	1.05029	.03203		1	1.41969	1.38498	1.05010	.03551	
	2	1.52984	1.46179	1.10333	.06307		2	1.45008	1.38088	1.10274	.06980	
	3	1.56325	1.46118	1.15941	.09319		3	1.48138	1.37779	1.15813	.10294	
	4	1.59788	1.46165	1.21882	.12243		4	1.51372	1.37573	1.21661	.13505	
	5	1.63385	1.46320	1.28190	.15086		5	1.54715	1.37464	1.27839	.16615	
	6	1.67130	1.46584	1.34907	.17853		6	1.58181	1.37460	1.34384	.19631	
	7	1.71037	1.46959	1.42075	.20547		7	1.61778	1.37540	1.41334	.22562	
	8	1.75122	1.47447	1.49749	.23174		8	1.65521	1.37744	1.48728	.25408	
	9	1.79404	1.48050	1.57988	.25738		9	1.69426	1.38037	1.56626	.28181	
	10	1.83906	1.48772	1.66862	.28242		10	1.73507	1.38431	1.65076	.30881	
	11	1.88651	1.49617	1.76495	.30690		11	1.77783	1.38936	1.74144	.33512	
	12	1.93668	1.50591	1.86864	.33085		12	1.82279	1.39545	1.83945	.36080	
	13	1.98991	1.51698	1.98205	.35430		13	1.87015	1.40266	1.94474	.38589	
	14	2.04657	1.52947	2.10617	.37728		14	1.92023	1.41102	2.05931	.41041	
	15	2.10714	1.54345	2.24267	.39983		15	1.97330	1.42055	2.18408	.43441	
	16	2.17214	1.55902	2.39362	.42196		16	2.02982	1.43138	2.32064	.45790	
	17	2.24223	1.57629	2.56152	.44370		17	2.09018	1.44353	2.47077	.48093	
	18	2.31820	1.59539	2.74953	.46507		18	2.15495	1.45708	2.63675	.50351	
	19	2.40103	1.61646	2.96164	.48610		19	2.22476	1.47212	2.82133	.52568	
	20	2.49191	1.63969	3.20294	.50680		20	2.30039	1.48878	3.02795	.54746	
	21	2.59237	1.66528	3.48010	.52720		21	2.38282	1.50715	3.26098	.56888	
	22	2.70435	1.69346	3.80196	.54731		22	2.47321	1.52740	3.52598	.58994	
	23	2.83381	1.72502	4.18819	.56715		23	2.57309	1.54968	3.83023	.61068	
	24	2.97391	1.75881	4.63254	.58675		24	2.68435	1.57449	4.18338	.63112	
	25	3.13997	1.79676	5.13317	.60612		25	2.80952	1.60116	4.59891	.65127	
	26	3.33394	1.83873	5.66488	.62526							
	27	3.56680	1.88545	6.33684	.64420							
	28	3.85318	1.93762	7.18996	.66295							
	29	4.21794	1.99614	8.28750	.68152							
	30	4.70573	2.06218	11.84954	.69994							
	31	5.40747	2.13722	15.70056	.71820							
	32	6.34844	2.22324	23.10298	.73634							
33	8.93919	2.32287	43.19541	.75435								
44	0	1.43955	1.43955	1.00000	0	47	0	1.43955	1.43955	1.00000	0	
	1	1.47017	1.43595	1.05016	.03315		1	1.46117	1.43245	1.09981	.03617	
	2	1.50175	1.43340	1.10300	.06524		2	1.48438	1.42994	1.16044	.07044	
	3	1.53439	1.43194	1.15878	.09634		3	1.50915	1.42751	1.22811	.10689	
	4	1.56816	1.43154	1.21777	.12651		4	1.53546	1.42516	1.30284	.14552	
	5	1.60320	1.43215	1.28033	.15581		5	1.56330	1.42289	1.38464	.18644	
	6	1.63960	1.43385	1.34680	.18429		6	1.59266	1.42070	1.47464	.22974	
	7	1.67754	1.43659	1.41765	.21202		7	1.62354	1.41859	1.57394	.27552	
	8	1.71712	1.44044	1.49329	.23900		8	1.65594	1.41656	1.68264	.32388	
	9	1.75855	1.44536	1.57435	.26532		9	1.69086	1.41461	1.80094	.37492	
	10	1.80201	1.45143	1.66148	.29101		10	1.72830	1.41274	1.92894	.42874	
	11	1.84775	1.45868	1.75542	.31608		11	1.76834	1.41094	2.06664	.48532	
	12	1.89597	1.46707	1.85710	.34062		12	1.81098	1.40921	2.21404	.54474	
	13	1.94702	1.47675	1.96754	.36461		13	1.85632	1.40754	2.37124	.60704	
14	2.00123	1.48774	2.08804	.37474	14	1.90446	1.40594	2.53834	.67332			





TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

Table with columns: theta (deg), beta (deg), Mb, Ma, Pa/Pb, Cp,ab/Cb, theta (deg), beta (deg), Mb, Ma, Pa/Pb, Cp,ab/Cb. The table is organized into two main sections, 50 and 51, with sub-sections for theta values 0 through 35 degrees.













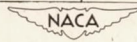






TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Concluded

$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_A$	$\frac{P_A}{P_b}$	$\frac{\Delta P_{A,b}}{q_b}$	$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_A$	$\frac{P_A}{P_b}$	$\frac{\Delta P_{A,b}}{q_b}$	
85	20	6.33047	0.44327	46.23224	1.61242	87	5	2.01291	0.58128	4.54750	1.25076	
	21	8.12762	.43674	76.31580	1.62878		6	2.25574	.54695	5.75347	1.33456	
86	0	1.00244	1.00244	1.00000	0	7	2.52995	.51836	7.28229	1.40171		
	1	1.15055	.87994	1.37020	.39951	8	2.84902	.49410	9.27715	1.45677		
	2	1.29591	.79401	1.78307	.66612	9	3.23409	.47324	12.00248	1.50275		
	3	1.44179	.72957	2.24675	.85680	10	3.72148	.45507	15.94670	1.54176		
	4	1.59085	.67901	2.77155	1.00000	11	4.38141	.43910	22.16816	1.57528		
	5	1.74555	.63805	3.37082	1.11156	12	5.37519	.42495	33.44909	1.60442		
	6	1.90850	.60405	4.06208	1.20098	13	7.20410	.41232	60.21636	1.62999		
	7	2.08259	.57529	4.86377	1.27428	14	13.26155	.40099	204.45147	1.65262		
	8	2.27139	.55060	5.82313	1.33551	88	0	1.00061	1.00061	1.00000	0	
	9	2.47941	.52914	6.97050	1.38744		1	1.29222	.79143	1.77909	.66653	
	10	2.71274	.51031	8.37701	1.43208		2	1.58355	.67540	2.75534	1.00000	
	11	2.97992	.49365	10.14282	1.47087		3	1.89445	.59221	4.01532	1.20025	
	12	3.29358	.47880	12.42734	1.50491		4	2.24475	.54437	5.70487	1.33388	
	13	3.67355	.46548	15.50082	1.53505		5	2.66202	.50254	8.09068	1.42944	
	14	4.15329	.45349	19.86017	1.56194		6	3.19358	.46933	11.71764	1.50122	
	15	4.79538	.44264	26.53113	1.58608		7	3.93752	.44219	17.89940	1.57114	
	16	5.73559	.43279	38.02639	1.60789		8	5.15650	.41950	30.81660	1.60196	
	17	7.35053	.42382	62.56201	1.62771		9	8.01593	.40020	74.70644	1.63870	
18	11.43133	.41564	151.54614	1.64580	89		0	1.00015	1.00015	1.00000	0	
87	0	1.00137	1.00137	1.00000			0	1	1.58174	.67450	2.75133	1.00000
	1	1.19742	.84687	1.50153			.49969	2	2.23822	.54284	5.67614	1.33347
	2	1.39022	.74753	2.08199			.79976	3	3.17002	.46702	11.55363	1.50031
	3	1.58658	.67690	2.76207	1.00000		4	5.03780	.41629	29.43369	1.60049	
4	1.79219	.62346	3.57035	1.14321								



SUPPLEMENT

NACA TECHNICAL NOTE NO. 1143

CHARTS FOR DETERMINING THE CHARACTERISTICS OF  
SHARP-NOSE AIRFOILS IN TWO-DIMENSIONAL  
FLOW AT SUPERSONIC SPEEDS

By H. Reese Ivey, George W. Stickle,  
and Alberta Schuettler

September 1947

Since Technical Note No. 1143 was completed, the need for a more extensive version of table I, "Values of Local Mach Number, Pressure Ratio, and Pressure Coefficient across Shock Waves," has become apparent. This table is now available in expanded form; and a copy of the expanded table is included in this supplement to supersede the original table I.

Errors in the original publication are as follows:

Page 11: The first sentence of the last paragraph should begin "Tables I and III . . ." instead of "Tables I and II . . . ."

Corrections in tables II and III are as follows:

Table II.--

$M_b$	$-\beta$	$M_a$
2.3	23°	3.4225
2.6	10°	3.0867
4.3	3°	4.5658
4.7	7°	5.4669
4.7	8°	5.5922
4.7	9°	5.7240
6.2	9°	7.9200

Table III.--

$M_b$	$-\beta$	$p_a/p_b$
1.4	14°	0.49071
4.7	24°	.02350

Figure 13.- Each vertical space along the scale label for pressure coefficient  $\Delta p/q$  should represent 0.125 instead of 20; thus, the vertical scale should appear:

-.500  
-.375  
-.250  
-.125  
0  
.125  
.250  
.375  
.500  
.625  
.750  
.875  
1.000









TABLE I - VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Continued

Table with columns for theta (deg), beta (deg), Mb, Ma, Pa/Pb, and delta Pa/b / qb. It is a multi-column data table providing aerodynamic parameters for shock waves.



TABLE I.-- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES -- Continued

Table with 12 columns: theta (deg), beta (deg), Mb, Ma, Pa/Pb, (Pa/Pb)0, theta (deg), beta (deg), Mb, Ma, Pa/Pb, (Pa/Pb)0. Data rows are grouped by theta values (46, 47, 48, 49, 50).

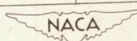


















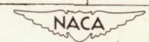
TABLE I.—VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND  
PRESSURE COEFFICIENT ACROSS SHOCK WAVES — Continued

Table with 12 columns: theta (deg), beta (deg), Mb, Ma, Pa/Pb, (Pb-a)/qb, theta (deg), beta (deg), Mb, Ma, Pa/Pb, (Pb-a)/qb. The table is organized into sections for theta = 76, 77, 78, 79, and 80, with corresponding beta values and Mach numbers.



TABLE I.- VALUES OF LOCAL MACH NUMBER, PRESSURE RATIO, AND PRESSURE COEFFICIENT ACROSS SHOCK WAVES - Concluded

$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$	$\theta$ (deg)	$\beta$ (deg)	$M_b$	$M_a$	$\frac{P_a}{P_b}$	$\frac{\Delta P_{a,b}}{q_b}$
85	20	6.33047	0.44327	46.23224	1.61242	87	5	2.01291	0.58128	4.54750	1.25076
	21	8.12762	.43674	76.31580	1.62878		6	2.25574	.54695	5.75347	1.33456
86	0	1.00244	1.00244	1.00000	0	7	2.52995	.51836	7.28029	1.40171	
	1	1.15055	.87994	1.37020	.39951	8	2.84902	.49410	9.27715	1.45677	
	2	1.29591	.79401	1.78307	.66612	9	3.23409	.47324	12.00248	1.50275	
	3	1.44179	.72957	2.24675	.85680	10	3.72148	.45507	15.94670	1.54176	
	4	1.59085	.67901	2.77155	1.00000	11	4.38141	.43910	22.16816	1.57528	
	5	1.74555	.63805	3.37082	1.11156	12	5.37519	.42495	33.44909	1.60442	
	6	1.90850	.60405	4.06208	1.20098	13	7.20410	.41232	60.21636	1.62999	
	7	2.08259	.57529	4.86377	1.27428	14	13.26155	.40099	204.45147	1.65262	
	8	2.27139	.55060	5.82313	1.33551	88	0	1.00061	1.00000	0	
	9	2.47941	.52914	6.97050	1.38744		1	1.29222	.79143	1.77909	.66653
	10	2.71274	.51031	8.37701	1.43208		2	1.58355	.67540	2.75534	1.00000
	11	2.97992	.49365	10.14282	1.47087		3	1.89445	.59921	4.01532	1.20025
12	3.29358	.47880	12.42734	1.50491	4	2.24475	.54437	5.70487	1.33388		
13	3.67355	.46548	15.50082	1.53505	5	2.66202	.50254	8.09068	1.42944		
14	4.15329	.45349	19.86017	1.56194	6	3.19358	.46933	11.71764	1.50122		
15	4.79538	.44264	26.53113	1.58608	7	3.93752	.44219	17.89940	1.55714		
16	5.73559	.43279	38.02639	1.60789	8	5.15650	.41950	30.81660	1.60196		
17	7.35053	.42382	62.56201	1.62771	9	8.01993	.40020	74.70644	1.63870		
18	11.43133	.41564	151.54614	1.64580	89	0	1.00015	1.00015	1.00000	0	
87	0	1.00137	1.00137	1.00000		0	1.58174	.67450	2.75133	1.00000	
	1	1.19742	.84687	1.50153		.49969	2	2.23822	.54284	5.67614	1.33347
	2	1.39022	.74753	2.08199		.79976	3	3.17002	.46702	11.55363	1.50031
	3	1.58658	.67690	2.76207	1.00000	4	5.03780	.41629	29.43369	1.60049	
4	1.79219	.62346	3.57035	1.14321							



CHARTS FOR DETERMINING THE CHARACTERISTICS OF  
SHARP-NOSE AIRFOILS IN TWO-DIMENSIONAL  
FLOW AT SUPERSONIC SPEEDS

By H. Reese Ivey, George W. Stickle,  
and Alberta Schuettler

SUMMARY

Solutions of the Hugoniot shock equations and Meyer expansion equations are plotted in such a manner as to permit the pressure distribution, the local Mach number, and the angles of shock waves on arbitrary sharp-nose airfoils at supersonic speeds to be obtained directly.

INTRODUCTION

Ackeret, in reference 1, gives a method for calculating the pressure distribution over thin, sharp, two-dimensional airfoils at supersonic speeds. This method, based on the theory of small disturbances, is only a first approximation and therefore is most accurate for thin airfoils.

The exact relationship for the pressure rise through a normal shock wave, as given by Hugoniot, is discussed in reference 2. According to reference 3, the corresponding relations which apply directly to the pressures on a straight surface of an airfoil immediately behind an oblique shock were obtained by Meyer as early as 1908. A discussion of Meyer's equations for the expansion of supersonic flow around an infinite corner is also given in reference 3. Frequently, interference exists between

shock and expansion waves caused by the intersection of two or more of these waves. When this intersection is close to the airfoil, as, for instance, when the airfoil has considerable curvature, the calculations yielded by the aforementioned equations are not exact.

It has been shown by Ferri (reference 4) that the equations for an oblique shock combined with the expansion equations give a close approximation to experimental results as reviewed in the section "Presentation of Figures" in this report. The use of the equations, however, involves long and difficult computations. The purpose of this paper is to give graphic solutions of these equations in a form suitable for rapid calculation. Because the size of the graphs limits their accuracy, tables are given from which computational graphs of much greater accuracy may be plotted. The relations given herein apply directly to a two-dimensional, or cylindrical, flow in which the transverse velocity is supersonic. As pointed out by Busemann (reference 5) they may be adapted to the case of oblique motion of the cylindrical airfoil by the addition of an arbitrary axial velocity. Thus, as in reference 5, the relations may be applied to the case of a swept-back airfoil lying ahead of the Mach lines, in which case the velocities and Mach numbers used in the calculation are those corresponding to the transverse component of the flight velocity. In case the airfoil is swept behind the Mach lines the flow will be of a different type as discussed in reference 6.

#### SYMBOLS

M	Mach number
p	static pressure
q	dynamic pressure
$\beta$	change in direction of flow (see fig. 1)
$\gamma$	ratio of specific heat at constant pressure to specific heat at constant volume = 1.4 for air
$\theta$	angle of shock wave relative to direction of flow before shock

$\nu$  angle around which the flow would have to expand from  $M = 1$  to the given local Mach number

$\rho$  density of gas

Subscripts:

a for conditions after a disturbance

b for conditions before a disturbance

n for local condition under consideration

o free stream

#### PRESENTATION OF FIGURES

A supersonic two-dimensional air flow around an airfoil may change its direction either by deflection or by expansion around a corner. In case the change in air-flow direction occurs by deflection, a shock wave is set up, and in case the change is by expansion, an expansion wave is set up. In either case, the change of state of the gas can be presented as a function of the local Mach number before the disturbance and the change in direction of the gas.

The equations from which the charts presented herein are derived are given in the appendix. Values of local Mach number, pressure ratio, and pressure coefficient across shock waves are given in table I. The local Mach numbers before and after expansions are presented in table II and the static relations across expansion waves, in table III. Table IV gives the pressure ratios based on free-stream dynamic pressure for various Mach numbers.

Before the method of determining pressure distribution, lift, drag, and moments may be discussed, a method of measuring the angles that cause expansions and shocks must be selected. Figure 1 shows the method used in the present paper for measuring angles causing expansions; figure 2 shows that for measuring angles causing shocks. The angle causing the disturbance is designated  $\beta$  in both cases;  $\beta$  is considered negative if the disturbance set up is an expansion wave and positive if it is a shock wave.



Figure 3 shows the manner in which the flow changes when the angles are made too large for the given speeds. If the angle causing the shock is too great, the shock wave separates from the airfoil surface. In figure 4 are given the maximum angles that may exist before the shock wave separates, calculated as the boundary condition between the region giving two solutions and the region for which no solution exists. If the trailing portion of the airfoil is too blunt, the flow may separate from the airfoil and leave a turbulent wake. Figure 3(b) shows that the expansion of the flow outside the wake is actually less than it would have been had it followed the surface. The pressure on the back of the airfoil does not decrease so much as it would if the flow failed to separate. The drags calculated if no separation is assumed will therefore be higher than those actually experienced.

The local Mach number after a disturbance (shock or expansion) is shown in figure 5 to be a function of the local Mach number before the disturbance and the angle causing the disturbance. For example, if a flow at a Mach number of 4.0 impinges on a surface set at  $5^\circ$  to that flow, a shock wave is set up behind which the local Mach number is 3.64, while the flow behind the shock wave is parallel to the surface. On the other hand, the same flow expanding around a  $5^\circ$  corner produces a local Mach number equal to 4.4 on the surface behind the expansion.

Figure 6 gives the ratio of static pressures across shock and expansion waves. For example, assume that a flow at a local Mach number of 4.0 is shocked by a surface slope change of  $\beta = 5^\circ$ . From figure 6 the pressure ratio across the shock is 1.61, which means that the pressure is much higher on the surface behind the shock than on the surface before it. If the flow at  $M = 4.0$  had expanded  $5^\circ$ , then the pressure would have dropped to 0.588. From this example it is seen that the  $5^\circ$  shock increased the pressure by 61 percent, whereas the  $5^\circ$  expansion decreased the pressure only 41 percent. Ackeret's method in reference 1 predicts equal changes in pressure for both the shock and the expansion, since it is based on small disturbances. Present results indicate, therefore, that angles as large as  $5^\circ$  require a more accurate approximation than that given by Ackeret.

The use of figures 5 and 6 can be demonstrated by solving for the local Mach numbers and pressures on the simple airfoil shown in figure 7. The coordinates of figures 5 and 6 are based on conditions before the disturbance and conditions after the disturbance. In figure 7 the conditions after one disturbance are noted to be the conditions before another disturbance. The numerical subscripts found in the symbols of figure 7 are to be associated, therefore, for use in the charts of figures 5 and 6, with the subscripts a and b, according to their relative positions with respect to the disturbance. The airfoil of figure 7 is a symmetrical, double-wedge airfoil having a  $2^\circ$  included angle at the leading and trailing edges. For use in this example the airfoil is at a positive angle of attack of  $3^\circ$  and is moving at a free-stream Mach number of 4.0. The pertinent angles as well as the conditions to be determined are shown on figure 7.

Enter figure 5 at  $M_0 = 4.0$  and  $\beta_1 = -2^\circ$ , and read off  $M_1 = 4.16$ . This Mach number is used to obtain  $M_2 = 4.33$ . Values for the lower surface of the airfoil are obtained in a similar manner. At coordinates of  $M_0 = 4.0$  and  $\beta_3 = 4^\circ$ ,  $M_3 = 3.70$ ; and, similarly, when the flow at  $M_3$  is expanded  $2^\circ$ ,  $M_4$  is found to be 3.84. A shock wave and an expansion wave are shown at the trailing edge; however, since these disturbances do not affect the pressures on the airfoil, they will be neglected, and  $M_1$ ,  $M_2$ ,  $M_3$ , and  $M_4$  are the only Mach numbers which are discussed.

The pressure ratios across the shock and expansion waves can be determined from figure 6. Enter figure 6 at  $M_0 = 4.0$  and an expansion angle of  $2^\circ$  and read

$$\frac{p_1}{p_0} = 0.817. \quad \text{At coordinates of } M_1 = 4.16 \quad \text{and} \quad \beta_2 = -2^\circ,$$

$$\frac{p_2}{p_1} \text{ is found to be } 0.809. \quad \text{Then}$$

$$\frac{p_2}{p_0} = \frac{p_1}{p_0} \frac{p_2}{p_1} = 0.817 \times 0.809 = 0.661$$

For the lower surface of the airfoil,  $p_3/p_0$  is found at coordinates  $M_0 = 4.0$ ,  $\beta_3 = 4^\circ$  to be  $\frac{p_3}{p_0} = 1.47$ .

At  $M_3 = 3.70$ ,  $\beta_4 = -2^\circ$ ,  $\frac{p_4}{p_3} = 0.829$ . Then

$$\frac{p_4}{p_0} = \frac{p_4}{p_3} \times \frac{p_3}{p_0} = 1.219$$

The pressure ratios  $\frac{p_a}{p_b}$  may be converted to local pressure coefficients  $\frac{\Delta p}{q_0}$  by the use of the plot given in figure 8. The results obtained for the local Mach numbers, pressure ratios, and pressure coefficients are illustrated in figure 9.

Once the pressure distribution is determined, the lift, drag, and moment coefficients can be obtained by integrating plots of the types given in figures 10, 11, and 12. The lift coefficient is obtained by integrating the projection of the airfoil pressure distribution on a plane parallel to free-stream direction. For the example airfoil at Mach number 4.0 and angle of attack  $3^\circ$ , the lift coefficient is 0.0540. The drag coefficient is found in the same manner except that the integration is over the projection of the airfoil pressure distribution on a plane perpendicular to free stream. The pressure drag coefficient for the example airfoil is 0.00315. The total section drag coefficient is the sum of the viscous and pressure-drag coefficients; for instance, if the viscous-drag coefficient is 0.0060, then the total drag coefficient is  $0.0060 + 0.00315 = 0.00915$ . The moment coefficient, obtained by integrating the elemental moments about the point desired, becomes 0.001112 when taken about the center for the example airfoil.

In the preceding examples step-by-step calculations were made along the airfoil, in which case the results obtained at any point are dependent on the accuracy of those at the preceding points. The results thus obtained on the rear of the airfoil may be subject to greater inaccuracies than are necessary. A method is consequently

given for determining the conditions behind each of a series of expansions independently of the conditions existing at intermediate points. For the example airfoil of figure 7, the free-stream flow was expanded around a  $2^\circ$  corner to obtain the conditions on the front of the upper surface and these conditions were then expanded around the second  $2^\circ$  corner to give the conditions on the rear of the upper surface. These last conditions, however, could have been found directly by referring the rear of the upper surface to the free-stream conditions and expanding through the total angle ( $4^\circ$ ) at once. Theoretically the results obtained are exactly the same regardless of which method is used, provided no shock waves are present between the end-points of the calculation. This method of adding angles does not apply when these are intermediate shock waves because of loss of total head in the shock wave.

If it is desired to calculate only the pressure distribution, it is not necessary to find  $M_1$ ,  $M_2$ ,  $M_4$ , or  $p_2/p_1$  for an airfoil similar to the type in the example given.

Figure 13 is taken from reference 4 to compare the experimental pressure distribution on an airfoil with the calculated distribution. Even though the wind-tunnel tests were of a very small model and although the airfoil is not of a type particularly suitable for calculations, the calculated and experimental values seem to compare favorably except for the region of separated flow near the upper trailing edge of the airfoil. The method of the present report is not exact for an airfoil of this type, which has considerable curvature along its entire length. The inaccuracy caused by the curvature, however, seems to be small. The thickness ratio and angle of attack of the example airfoil are somewhat higher than those for which the method is recommended. Reference 4 explains the separated region of flow.

Langley Memorial Aeronautical Laboratory  
National Advisory Committee for Aeronautics  
Langley Field, Va., April 4, 1946

## APPENDIX

## METHOD OF ANALYSIS

## Shock Waves

Supersonic air flow about an airfoil may be said to consist of expansions and shocks. Reference 2 mentions the fact that a change in entropy occurs through a shock wave. Three conditions are shown, however, to apply to the velocities, pressures, and densities at the two sides of the shock wave, namely:

(a) Continuity of mass

(b) Balance between pressure difference and change of momentum

(c) Conservation of energy

These conditions lead to the three basic equations:

or

$$\left. \begin{aligned} \frac{\rho_a}{\rho_b} &= \frac{\tan \theta}{\tan (\theta - \beta)} \\ \frac{\Delta p}{\rho_a} &= \frac{\sin \beta}{\sin \theta \times \cos (\theta - \beta)} \end{aligned} \right\} (1)$$

$$\left. \begin{aligned} \frac{\Delta p}{q_b} &= 2 \sin^2 \theta \frac{\Delta p}{\rho_a} \\ &= \frac{2 \sin \beta \times \sin \theta}{\cos (\theta - \beta)} \end{aligned} \right\} (2)$$

$$\frac{\Delta p}{\Delta \rho} = \gamma \frac{p_b + \frac{\Delta p}{2}}{\rho_b + \frac{\Delta \rho}{2}} \quad (3)$$

Then, by use of the relation

$$\frac{\Delta p}{\rho_b} = \frac{2}{\gamma M_b^2} \left( \frac{p_a}{p_b} - 1 \right) \quad (4)$$

it follows that

$$\frac{p_a}{p_b} = \frac{\gamma M_b^2 \sin \beta \sin \theta}{\cos (\theta - \beta)} + 1 \quad (5)$$

$$\frac{1}{M_b^2} = \sin^2 \theta - \frac{\gamma + 1}{2} \frac{\sin \beta \sin \theta}{\cos (\theta - \beta)} \quad (6)$$

$$M_a = M_b \frac{\cos \theta}{\cos (\theta - \beta)} \sqrt{\frac{p_b \rho_a}{p_a \rho_b}} \quad (7)$$

By substitution of arbitrary values of  $\theta$  and  $\beta$  in equations (1) and (6), the corresponding values of density ratio across the shock and Mach number before the shock are obtained. If the simultaneous values of  $\theta$ ,  $\beta$ , and  $M_b$  are used with equations (5) and (7), the pressure ratio across the shock and the Mach number after the shock are obtained.

Figure 14 shows the angle of the shock wave as a function of the Mach number before the shock  $M_b$  and the angle defining the change in direction of the flow  $\beta$ .

The pressure ratios and the Mach numbers after the shock have already been discussed for figures 5 and 6. Use of the ratio of pressure after any shock wave to free-stream static pressure, together with free-stream Mach number in equation (4), makes possible the determination of the pressure coefficient behind that shock wave. Figure 8 has shown the graph for converting pressure ratios to pressure coefficients.

#### Expansion Waves

The flow after the shock wave may be considered adiabatic as long as the flow is expanding. By the use of such flow conditions, the velocities, densities, and pressures may be calculated. Experimentally some trouble is encountered when extremely large angles of expansion are used. The flow may break down and form a turbulent wake of somewhat higher static pressure than might be expected if the flow had continued to expand around the corner.

Reference 3 considers that a flow at a Mach number of 1 expands around some angle  $v$  and reaches a higher Mach number  $M$  defined by the relation

$$v = \sqrt{\frac{\gamma + 1}{\gamma - 1}} \tan^{-1} \left( \sqrt{M^2 - 1} \sqrt{\frac{\gamma - 1}{\gamma + 1}} \right) - \cos^{-1} \frac{1}{M} \quad (8)$$

By expanding around an angle  $v_b$ , the flow reaches a Mach number  $M_b$ ; and by expanding around some larger angle  $v_a$ , the flow reaches some higher Mach number  $M_a$ . A flow at the first Mach number  $M_b$  can then reach the higher Mach number  $M_a$  by expanding around the small angle,

$$-\beta = v_a - v_b \quad (9)$$

Equations (8) and (9) serve as the basis for calculating the expansion lines in figure 5 showing the variation of local Mach number with change in surface slope.

Another equation derived from the work of reference 3 gives the pressure ratio across expansion waves as

$$\frac{p_a}{p_b} = \left[ \frac{2 + (\gamma - 1) M_b^2}{2 + (\gamma - 1) M_a^2} \right]^{\frac{\gamma}{\gamma - 1}} \quad (10)$$

From equations (8), (9), and (10), it is possible to calculate the part of figure 6 that gives the pressure ratio across expansion waves as a function of the local Mach number before the expansion and the change in surface slopes.

The figures shown in this report, because of their limited size, may not be accurate enough for routine calculations. It may be desirable to plot the graphs to a larger scale before using them. For this reason the values are listed in tabular form for the main graphs.

Tables I and III should be accurate to all the figures shown, but table II may not be exact in the last figure since the expansion calculations required graphical interpolation between very close computed points.



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TABLE III.- STATIC PRESSURE RATIO ACROSS EXPANSION WAVES

		Ratio of static pressure after to static pressure before expansion, $p_a/p_b$														
$M_b$	$\beta$	0°	1°	2°	3°	4°	5°	6°	7°	8°	9°	10°	11°	12°	13°	14°
1.0	1	0.90724	0.85100	0.80486	0.76192	0.72329	0.68896	0.65588	0.62488	0.59484	0.56608	0.53921	0.51325	0.48875	0.46512	
1.1	1	.94222	.89109	.84500	.80315	.76474	.72819	.69275	.65930	.62796	.59815	.56975	.54234	.51571	.49054	
1.2	1	.94939	.90167	.85851	.81779	.77827	.74080	.70538	.67194	.63907	.60823	.57938	.55105	.52394	.49764	
1.3	1	.95151	.90630	.86285	.82202	.78284	.74496	.70863	.67461	.64159	.61019	.58005	.55075	.52301	.49658	
1.4	1	.95252	.90701	.86361	.82165	.78169	.74383	.70671	.67176	.63872	.60612	.57570	.54610	.51755	.49071	
1.5	1	.95205	.90613	.86205	.81855	.77832	.73967	.70135	.66630	.63249	.59908	.56706	.53637	.50922	.48197	
1.6	1	.95145	.90424	.85851	.81561	.77403	.73369	.69275	.65180	.61172	.58111	.55086	.52086	.49339	.46723	
1.7	1	.95003	.90151	.85444	.81159	.76833	.72739	.68686	.64618	.60572	.56544	.52533	.48533	.44533	.41681	
1.8	1	.94508	.89979	.85116	.80630	.76189	.72019	.68066	.64211	.60376	.56566	.52779	.48997	.45226	.41468	
1.9	1	.94568	.89744	.84615	.79856	.75373	.71173	.67047	.62925	.58816	.54710	.50618	.46531	.42448	.38368	
2.0	1	.94504	.89218	.84172	.79217	.74492	.70301	.66159	.62203	.58523	.54831	.51391	.48134	.44600	.42098	
2.1	1	.94353	.88893	.83644	.78776	.74007	.69517	.65234	.61110	.57282	.53622	.50140	.46847	.43396	.40484	
2.2	1	.94169	.88533	.83124	.78061	.73139	.68604	.64269	.60121	.56152	.52471	.48990	.45533	.42137	.38768	
2.3	1	.94051	.88174	.82668	.77403	.72460	.67861	.63479	.59245	.55143	.51223	.47630	.44266	.41117	.38111	
2.4	1	.93750	.87832	.82216	.76729	.71632	.66824	.62225	.57905	.53839	.50097	.46415	.42802	.39324	.36028	
2.5	1	.93553	.87396	.81507	.75970	.70781	.65891	.61160	.56795	.52732	.48915	.45188	.41744	.38531	.35531	
2.6	1	.93326	.87003	.80943	.75322	.69983	.64981	.60185	.55743	.51641	.47724	.44099	.40756	.37683	.34811	
2.7	1	.93040	.86544	.80426	.74619	.69245	.64269	.59257	.54717	.50601	.46763	.43126	.39742	.36523	.33499	
2.8	1	.92819	.86210	.79888	.73977	.68522	.63118	.58209	.53716	.49476	.45304	.41210	.37267	.33499	.31853	
2.9	1	.92725	.85754	.79430	.73520	.67538	.62285	.57313	.52582	.48269	.44310	.40476	.36792	.33288	.30826	
3.0	1	.92417	.85269	.78891	.72544	.66790	.61419	.56305	.51517	.47200	.43145	.39323	.35841	.32612	.29566	
3.1	1	.92232	.85032	.78316	.71897	.65964	.60524	.55329	.50543	.46134	.41998	.38213	.34740	.31438	.28476	
3.2	1	.92014	.84621	.77736	.71173	.65139	.59522	.54330	.49504	.45083	.40936	.37115	.33610	.30338	.27400	
3.3	1	.91833	.84237	.77205	.70478	.64379	.58651	.53438	.48518	.44040	.39926	.36075	.32543	.29305	.26357	
3.4	1	.91582	.83832	.76596	.69828	.63618	.57808	.52522	.47528	.43051	.38863	.34974	.31502	.28254	.25312	
3.5	1	.91317	.83493	.75984	.69097	.62874	.56930	.51495	.46607	.42036	.37797	.33972	.30465	.27259	.24355	
3.6	1	.91113	.83069	.75433	.68436	.62068	.56032	.50620	.45613	.41025	.36744	.32991	.29456	.26275	.23390	
3.7	1	.90881	.82674	.74877	.67711	.61325	.55226	.49724	.44679	.40044	.35836	.32194	.28793	.25540	.22661	
3.8	1	.90695	.82299	.74320	.67034	.60509	.54325	.48882	.43714	.39091	.34883	.31200	.27838	.24620	.21699	
3.9	1	.90506	.81916	.73702	.66478	.59735	.53577	.47962	.42832	.38192	.33987	.30217	.26828	.23550	.20569	
4.0	1	.90314	.81531	.73244	.65834	.58945	.52725	.47074	.41923	.37201	.32936	.29171	.25720	.22430	.19486	
4.1	1	.90057	.81090	.72705	.65172	.58125	.51932	.46191	.41008	.36262	.32086	.28305	.24848	.21580	.18619	
4.2	1	.89917	.80717	.72102	.64463	.57215	.51084	.45159	.40159	.35466	.31140	.27425	.24023	.20882	.18261	
4.3	1	.89622	.80232	.71563	.63879	.56671	.50343	.44472	.39244	.34578	.30060	.26574	.23192	.20192	.17510	
4.4	1	.89351	.79861	.71027	.63205	.55989	.49552	.43644	.38449	.33726	.29518	.25736	.22364	.19449	.16781	
4.5	1	.89199	.79481	.70543	.62568	.55343	.48754	.42920	.37624	.32879	.28733	.24945	.21636	.18657	.16062	
4.6	1	.89018	.79125	.70037	.61909	.54603	.47970	.42124	.36783	.32078	.27891	.24155	.20871	.17950	.15377	
4.7	1	.88755	.78643	.69482	.61275	.53871	.47264	.41259	.36009	.31265	.27114	.23395	.20088	.17251	.14731	
4.8	1	.88735	.78281	.69011	.60634	.53206	.46677	.40538	.35236	.30495	.26326	.22659	.19111	.16576	.14109	
4.9	1	.88360	.77862	.68548	.60051	.52542	.45755	.39757	.34465	.29765	.25590	.21926	.18731	.15932	.13488	
5.0	1	.88232	.77476	.67988	.59414	.51803	.45026	.39045	.33680	.28954	.24859	.21226	.18055	.15288	.12895	
5.1	1	.87976	.77103	.67527	.58794	.51132	.44357	.38317	.32662	.28274	.24143	.20543	.17404	.14681	.12333	
5.2	1	.87740	.76714	.67115	.58215	.50508	.43609	.37504	.32215	.27527	.23426	.19867	.16774	.14096	.11798	
5.3	1	.87542	.76370	.66640	.57631	.49959	.42890	.36684	.31488	.26826	.22755	.19216	.16145	.13529	.11262	
5.4	1	.87294	.75993	.65977	.56705	.49044	.41878	.35623	.30310	.26122	.22099	.18588	.15565	.12955	.10758	
5.5	1	.87136	.75556	.65448	.56163	.48519	.41336	.35028	.30087	.25470	.21155	.17966	.14997	.12441	.10268	
5.6	1	.86885	.75213	.64936	.55561	.47879	.40655	.34335	.29219	.24786	.20803	.17380	.14435	.11928	.09798	
5.7	1	.86733	.74837	.64397	.54960	.47233	.40012	.33710	.28736	.24146	.20187	.16794	.13901	.11288	.09350	
5.8	1	.86512	.74459	.63897	.54370	.46638	.39348	.33029	.28152	.23524	.19573	.16220	.13379	.10962	.08913	
5.9	1	.86329	.74111	.63395	.53779	.45961	.38673	.32378	.27463	.22910	.19003	.15677	.12862	.10488	.08502	
6.0	1	.86052	.73744	.62939	.53145	.45339	.38243	.32103	.26819	.22294	.18442	.15142	.12372	.10046	.08101	
6.1	1	.85855	.73377	.62410	.52498	.44757	.37593	.31450	.26209	.21702	.17863	.14639	.11899	.09613	.07720	
6.2	1	.85688	.72976	.61939	.52029	.44345	.37131	.30931	.25695	.21091	.17340	.14131	.11550	.09211	.07353	
6.3	1	.85466	.72674	.61557	.51581	.43959	.36718	.30520	.25316	.20631	.16979	.13828	.10996	.08791	.06933	
6.4	1	.85222	.72277	.61019	.51029	.43497	.36252	.30052	.24830	.20016	.16284	.13153	.10583	.08424	.06664	
6.5	1	.84960	.71889	.60548	.50586	.43079	.35836	.29636	.24434	.19641	.15796	.12709	.10164	.08059	.06315	
6.6	1	.84770	.71516	.60082	.50132	.42640	.35434	.29234	.23991	.19222	.15305	.12262	.09761	.07700	.06031	
6.7	1	.84603	.71282	.59702	.49761	.42370	.35192	.28992	.23719	.18919	.14963	.11984	.09373	.07364	.05731	
6.8	1	.84463	.70974	.59317	.49381	.42011	.34856	.28676	.23352	.18511	.14517	.11497	.08839	.06873	.05285	
6.9	1	.84168	.70778	.58886	.48961	.41612	.34484	.28304	.22939	.18111	.13989	.10936	.08311	.06381	.04831	
7.0	1	.83952	.70133	.58302	.48415	.41086	.33724	.27567	.22166	.16931	.13449	.10610	.08283	.06125	.04614	
7.1	1	.83705	.69763	.57789	.47994	.40666	.33302	.27157	.21718	.16475	.13036	.10222	.07961	.05761	.04294	
7.2	1	.83539	.69477	.57327	.47545	.40217	.32840	.26705	.21270	.16009	.12612	.09854	.07629	.05484	.04026	
7.3	1	.83296	.69014	.56923	.46625	.39297	.32019	.25894	.20479	.15157	.11822	.09493	.07313	.05214	.03796	
7.4	1	.83026	.68692	.56438	.46118	.38737	.31489	.25374	.20011	.14643	.11355	.08806	.06715	.04744	.03399	
7.5	1	.83125	.68388	.55986	.45621	.38290	.31042	.24927	.19564	.14201	.10944	.08388	.06273	.04362	.03062	
7.6	1	.82714	.68020	.55541	.45199	.37864	.30616	.24501	.19138	.13774	.10517	.07961	.05816	.03962	.02712	
7.7	1	.82479	.67668	.55118	.44633	.37402	.30154	.24039	.18676	.13312	.10055	.07500	.05385	.03572	.02372	
7.8	1	.82223	.67242	.54658	.44153	.36908	.29452	.23387	.17924	.12560	.09303	.06747	.04672	.02902	.01722	
7.9	1	.82043	.66952	.54279	.43714	.36426	.28960	.22895	.17432	.12068	.08811	.06255	.04180	.02450	.01300	
8.0	1	.81816	.66627	.53827	.43204	.35944	.28478	.22413	.16950	.11586	.08329	.05773	.03708	.02028	.00918	

TABLE III.-- STATIC PRESSURE RATIO ACROSS EXPANSION WAVES - Concluded

M <sub>1</sub> \ β	Ratio of static pressure after to static pressure before expansion, P <sub>a</sub> /P <sub>b</sub>															
	15°	16°	17°	18°	19°	20°	21°	22°	23°	24°	25°	26°	27°	28°	29°	30°
1.0	0.11233	0.12075	0.13099	0.17981	0.36053	0.31210	0.32450	0.30768	0.29152	0.27610	0.26136	0.24727	0.23324	0.22024	0.20805	0.19690
1.1	.16646	.11333	.12102	.59970	.37923	.35996	.31101	.32335	.30572	.28941	.27384	.25882	.24421	.23017	.21711	.20436
1.2	.17264	.14874	.12587	.40358	.38232	.34216	.31272	.32416	.30680	.28978	.27359	.25786	.24293	.22898	.21563	.20297
1.3	.17071	.14602	.12247	.40011	.37658	.35814	.33858	.31975	.30180	.28453	.26818	.25256	.23750	.22299	.20941	.19649
1.4	.16477	.13970	.11572	.39283	.37099	.35021	.33009	.31119	.29300	.27559	.25909	.24323	.22850	.21430	.20114	.18800
1.5	.15535	.12962	.10604	.38257	.36062	.33989	.31972	.30032	.28234	.26522	.24860	.23285	.21823	.20427	.19078	.17732
1.6	.14333	.11877	.99459	.37117	.34903	.32778	.30802	.28883	.27059	.25358	.23670	.22110	.20585	.19151	.17755	.16488
1.7	.13239	.10652	.86151	.35782	.33620	.31564	.29525	.27590	.25773	.24050	.22427	.20900	.19452	.18099	.16830	.15588
1.8	.11953	.93975	.86878	.34512	.32323	.30233	.28179	.26287	.24482	.22801	.21200	.19680	.18223	.16959	.15677	.14525
1.9	.10659	.80858	.85578	.33209	.30950	.28869	.26818	.24971	.23228	.21535	.19965	.18403	.17125	.15808	.14596	.13417
2.0	.99310	.86657	.81170	.31813	.29622	.27521	.25546	.23696	.21947	.20305	.18768	.17304	.15955	.14675	.13497	.12354
2.1	.98967	.85331	.82865	.30531	.28274	.26204	.24225	.22372	.20693	.19082	.17614	.16244	.14924	.13714	.12567	.11461
2.2	.96662	.83886	.81514	.29186	.26939	.24888	.22903	.21158	.19585	.18002	.16519	.15081	.13766	.12548	.11473	.10465
2.3	.95333	.82638	.80188	.27901	.25707	.23679	.21739	.19979	.18311	.16767	.15339	.13978	.12711	.11604	.10542	.09579
2.4	.93998	.81363	.78907	.26614	.24415	.22407	.20581	.18804	.17180	.15691	.14289	.12975	.11792	.10682	.09665	.08740
2.5	.92745	.80091	.77624	.25328	.23206	.21256	.19370	.17606	.16095	.14604	.13261	.12003	.10867	.09802	.08833	.07950
2.6	.91499	.78858	.76382	.24165	.22059	.20092	.18289	.16617	.15059	.13633	.12329	.11124	.10011	.08993	.08069	.07221
2.7	.90285	.77667	.75180	.22999	.20928	.19003	.17210	.15584	.14094	.12703	.11426	.10261	.09212	.08243	.07352	.06565
2.8	.89113	.76498	.74008	.21895	.19838	.17945	.16202	.14627	.13154	.11800	.10584	.09469	.08433	.07530	.06696	.05929
2.9	.87951	.75341	.72871	.20870	.18825	.16925	.15235	.13666	.12253	.10969	.09788	.08700	.07745	.06859	.06070	.05321
3.0	.86804	.74267	.71819	.19758	.17765	.15972	.14300	.12787	.11429	.10167	.09047	.08009	.07088	.06255	.05496	.04828
3.1	.85671	.73177	.70748	.18710	.16717	.15046	.13416	.11977	.10634	.09415	.08338	.07352	.06467	.05666	.04976	.04332
3.2	.84546	.72081	.69671	.17688	.15738	.14117	.12533	.11171	.09872	.08725	.07680	.06750	.05905	.05194	.04534	.03928
3.3	.83423	.70980	.68580	.16689	.14769	.13171	.11619	.10291	.09017	.07912	.06906	.06017	.05197	.04477	.03857	.03282
3.4	.82303	.69870	.67480	.15702	.13809	.12245	.10727	.09450	.08240	.07180	.06219	.05357	.04535	.03815	.03215	.02660
3.5	.81183	.68740	.66360	.14738	.12869	.11331	.09834	.08593	.07480	.06466	.05554	.04684	.03874	.03174	.02594	.02060
3.6	.80063	.67620	.65250	.13802	.11959	.10441	.08964	.07749	.06660	.05683	.04768	.03908	.03118	.02438	.01878	.01360
3.7	.78943	.66510	.64150	.12895	.11079	.09571	.08114	.07013	.05944	.04999	.04100	.03250	.02460	.01780	.01230	.00730
3.8	.77823	.65390	.63040	.12008	.10211	.08713	.07276	.06185	.05154	.04244	.03384	.02564	.01794	.01134	.00604	.00134
3.9	.76703	.64270	.61930	.11142	.09369	.07881	.06464	.05383	.04380	.03490	.02670	.01880	.01130	.00580	.00070	.00000
4.0	.75583	.63150	.60820	.10297	.08549	.07071	.05674	.04593	.03600	.02720	.01890	.01130	.00580	.00070	.00000	.00000
4.1	.74463	.62030	.59700	.09472	.07749	.06281	.04894	.03813	.02820	.01940	.01130	.00580	.00070	.00000	.00000	.00000
4.2	.73343	.60910	.58580	.08667	.07059	.05591	.04204	.03123	.02130	.01250	.00440	.00000	.00000	.00000	.00000	.00000
4.3	.72223	.59790	.57460	.07882	.06384	.04916	.03529	.02448	.01455	.00575	.00000	.00000	.00000	.00000	.00000	.00000
4.4	.71103	.58670	.56340	.07117	.05639	.04171	.02784	.01703	.00823	.00000	.00000	.00000	.00000	.00000	.00000	.00000
4.5	.70003	.57550	.55220	.06372	.04904	.03436	.02049	.00968	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
4.6	.68903	.56430	.54100	.05647	.04189	.02721	.01334	.00253	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
4.7	.67803	.55310	.52980	.04942	.03484	.02016	.00629	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
4.8	.66703	.54190	.51860	.04257	.02799	.01331	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
4.9	.65603	.53070	.50740	.03592	.02114	.00646	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.0	.64503	.51950	.49620	.02947	.01445	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.1	.63403	.50830	.48500	.02312	.00810	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.2	.62303	.49710	.47380	.01687	.00175	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.3	.61203	.48590	.46260	.01072	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.4	.60103	.47470	.45140	.00467	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.5	.59003	.46350	.44020	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.6	.57903	.45230	.42900	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.7	.56803	.44110	.41780	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.8	.55703	.42990	.40660	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
5.9	.54603	.41870	.39540	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.0	.53503	.40750	.38420	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.1	.52403	.39630	.37300	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.2	.51303	.38510	.36180	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.3	.50203	.37390	.35060	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.4	.49103	.36270	.33940	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.5	.48003	.35150	.32820	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.6	.46903	.34030	.31700	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.7	.45803	.32910	.30580	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.8	.44703	.31790	.29460	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
6.9	.43603	.30670	.28340	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
7.0	.42503	.29550	.27220	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
7.1	.41403	.28430	.26100	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000
7.2	.40303	.27310	.24980	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000	.00000

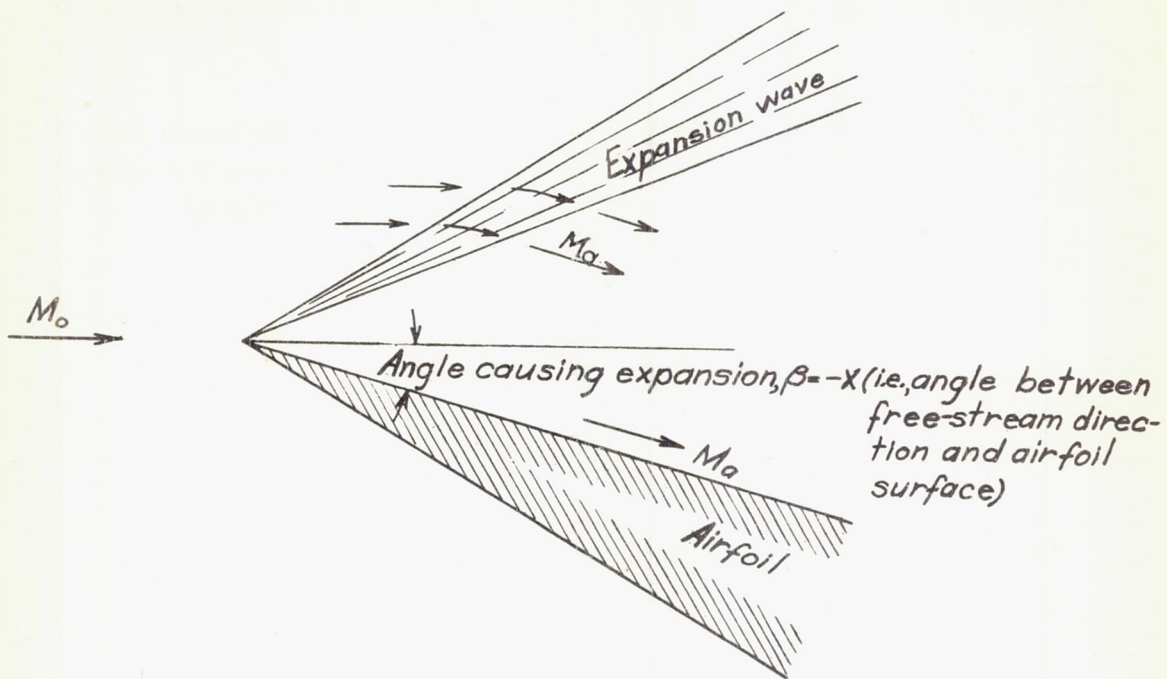
0235

TABLE IV.- PRESSURE COEFFICIENT BASED ON FREE-STREAM DYNAMIC PRESSURE

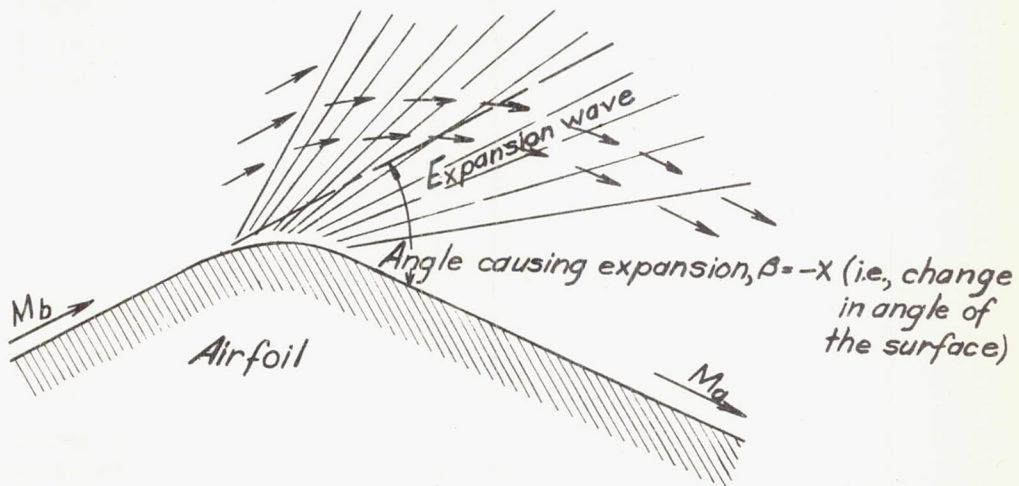
		Pressure coefficient, $\Delta p_n/q_\infty$							
		1.0	1.25	1.5	1.75	2.0	2.5	3.0	3.5
M <sub>0</sub> P <sub>n</sub> /P <sub>0</sub>	0	-1.42857	-0.91429	-0.63492	-0.47579	-0.35714	-0.22857	-0.15873	-0.11662
	.05	-1.35714	-.86857	-.60317	-.45200	-.33929	-.21714	-.15079	-.11079
	.1	-1.28571	-.82286	-.57143	-.42821	-.32143	-.20571	-.14286	-.10496
	.2	-1.14286	-.73143	-.50794	-.38064	-.28571	-.18286	-.12698	-.093294
	.3	-1.00000	-.64000	-.44144	-.33306	-.25000	-.16000	-.11111	-.081633
	.4	-.85714	-.54857	-.38095	-.28548	-.21429	-.13714	-.095238	-.069971
	.5	-.71429	-.45714	-.31746	-.23790	-.17857	-.11429	-.079365	-.058309
	.6	-.57143	-.36571	-.25397	-.19032	-.14286	-.091429	-.063492	-.046647
	.7	-.42857	-.27429	-.19048	-.14274	-.10714	-.068571	-.047619	-.034985
	.8	-.28571	-.18286	-.12698	-.095159	-.071429	-.045714	-.031746	-.023324
	.9	-.14286	-.091429	-.063492	-.047579	-.035714	-.022857	-.015873	-.011662
1.0	0	0	0	0	0	0	0	0	
1.1	.14286	.091429	.063492	.047579	.035714	.022857	.015873	.011662	
1.2	.28571	.18286	.12698	.095159	.071429	.045714	.031746	.023324	
1.3	.42857	.27429	.19048	.14274	.10714	.068571	.047619	.034985	
1.4	.57143	.36571	.25397	.19032	.14286	.091429	.063492	.046647	
1.5	.71429	.45714	.31746	.23790	.17857	.11429	.079365	.058309	
1.6	.85714	.54857	.38095	.28548	.21429	.13714	.095238	.069971	
1.7	1.00000	.64000	.44144	.33306	.25000	.16000	.11111	.081633	
1.8	1.14286	.73143	.50794	.38064	.28571	.18286	.12698	.093294	
1.9	1.28571	.82286	.57143	.42821	.32143	.20571	.14286	.10496	
2.0	1.42857	.91429	.63492	.47579	.35714	.22857	.15873	.11662	
2.2	1.71429	1.09714	.76190	.57095	.42857	.27429	.19048	.13994	
2.4	2.00000	1.28000	.88889	.66611	.50000	.32000	.22222	.16327	
2.6	2.28571	1.46286	1.01587	.76127	.57143	.36571	.25397	.18659	
2.8	2.57143	1.64571	1.14286	.85643	.64286	.41143	.28571	.20991	
3.0	2.85714	1.82857	1.26984	.95159	.71429	.45714	.31746	.23324	
3.2	3.14286	2.01143	1.39683	1.04675	.78571	.50286	.34921	.25656	
3.4	3.42857	2.19429	1.52381	1.14191	.85714	.54857	.38095	.27988	
3.6	3.71429	2.37714	1.65079	1.23706	.92857	.59429	.41270	.30321	
3.8	4.00000	2.56000	1.77778	1.33222	1.00000	.64000	.44144	.32653	
4.0	4.28571	2.74286	1.90476	1.42738	1.07143	.68571	.47619	.34985	
4.2	4.57143	2.92571	2.03175	1.52254	1.14286	.73143	.50794	.37318	
4.4	4.85714	3.10857	2.15873	1.61770	1.21429	.77714	.53968	.39650	
4.6	5.14286	3.29143	2.28571	1.71286	1.28571	.82286	.57143	.41983	
4.8	5.42857	3.47429	2.41270	1.80802	1.35714	.86857	.60317	.44315	
5.0	5.71429	3.65714	2.53968	1.90318	1.42857	.91429	.63492	.46647	
5.2	6.00000	3.84000	2.66667	1.99833	1.50000	.96000	.66667	.48980	
5.4	6.28511	4.02286	2.79365	2.09349	1.57143	1.00571	.69841	.51312	
5.6	6.57113	4.20571	2.92063	2.18865	1.64286	1.05143	.73016	.53644	
5.8	6.85714	4.38857	3.04762	2.28381	1.71429	1.09714	.76190	.55977	
6.0	7.14286	4.57143	3.17460	2.37897	1.78571	1.14286	.79365	.58309	
6.2	7.42857	4.75429	3.30159	2.47413	1.85714	1.18857	.82540	.60641	
6.4	7.71429	4.93714	3.42857	2.56929	1.92857	1.23429	.85714	.62974	
6.6	8.00000	5.12000	3.55556	2.66445	2.00000	1.28000	.88889	.65306	
6.8	8.28571	5.30286	3.68254	2.75960	2.07143	1.32571	.92063	.67638	
7.0	8.57143	5.48571	3.80952	2.85476	2.14286	1.37143	.95238	.69971	
7.2	8.85714	5.66857	3.93651	2.94992	2.21429	1.41714	.98413	.72303	
7.4	9.14286	5.85143	4.06349	3.04508	2.28571	1.46286	1.01587	.74636	
7.6	9.42857	6.03428	4.19048	3.14024	2.35714	1.50857	1.04762	.76968	
7.8	9.71429	6.21714	4.31746	3.23540	2.42857	1.55429	1.07936	.79300	
8.0	10.00000	6.40000	4.44444	3.33056	2.50000	1.60000	1.11111	.81633	
8.2	10.28571	6.58286	4.57143	3.42572	2.57143	1.64571	1.14286	.83965	
8.4	10.57143	6.76571	4.69841	3.52087	2.64286	1.69143	1.17460	.86297	
8.6	10.85714	6.94857	4.82541	3.61603	2.71429	1.73714	1.20635	.88630	
8.8	11.14286	7.13143	4.95238	3.71119	2.78571	1.78286	1.23809	.90962	
9.0	11.42857	7.31428	5.07936	3.80635	2.85714	1.82857	1.26984	.93294	
9.2	11.71429	7.49714	5.20635	3.90151	2.92857	1.87429	1.30159	.95627	
9.4	12.00000	7.68000	5.33333	3.99667	3.00000	1.92000	1.33333	.97959	
9.6	12.28571	7.86286	5.46032	4.09183	3.07143	1.96571	1.36508	1.00292	
9.8	12.57143	8.04571	5.58730	4.18699	3.14286	2.01143	1.39682	1.02624	
10.0	12.85714	8.22857	5.71429	4.28215	3.21429	2.05714	1.42857	1.04956	

TABLE IV.- PRESSURE COEFFICIENT BASED ON FREE-STREAM DYNAMIC PRESSURE - Concluded

		Pressure coefficient, $\Delta p_n/q_\infty$								
$M_0$		4.0	4.5	5.0	5.5	6.0	6.5	7.0	7.5	8.0
$P_n/P_0$										
0		-0.089286	-0.070547	-0.057143	-0.047226	-0.039683	-0.033812	-0.029155	-0.025397	-0.022321
.05		-.081821	-.067019	-.054286	-.044864	-.037698	-.032122	-.027697	-.024127	-.021205
.1		-.080357	-.063492	-.051429	-.042503	-.035714	-.030431	-.026239	-.022857	-.020089
.2		-.071429	-.056437	-.045714	-.037780	-.031746	-.027050	-.023324	-.020317	-.017857
.3		-.062500	-.049383	-.040000	-.033058	-.027778	-.023669	-.020408	-.017778	-.015625
.4		-.053571	-.042328	-.034286	-.028335	-.023810	-.020287	-.017493	-.015238	-.013393
.5		-.044643	-.035273	-.028571	-.023613	-.019841	-.016906	-.014577	-.012698	-.011161
.6		-.035714	-.028219	-.022857	-.018890	-.015873	-.013525	-.011662	-.010159	-.0089286
.7		-.026786	-.021164	-.017143	-.014168	-.011905	-.010144	-.0087464	-.0076190	-.0066964
.8		-.017857	-.014409	-.011429	-.0094451	-.0079365	-.0067625	-.0058309	-.0050794	-.0044643
.9		-.0089286	-.0070547	-.0057143	-.0047226	-.0039683	-.0033812	-.0029155	-.0025397	-.0022321
1.0	0	0	0	0	0	0	0	0	0	0
1.1		.0089286	.0070547	.0057143	.0047226	.0039683	.0033812	.0029155	.0025397	.0022321
1.2		.017857	.014409	.011429	.0094451	.0079365	.0067625	.0058309	.0050794	.0044643
1.3		.026786	.021164	.017143	.014168	.011905	.010144	.0087464	.0076190	.0066964
1.4		.035714	.028219	.022857	.018890	.015873	.013525	.011662	.010159	.0089286
1.5		.044643	.035273	.028571	.023613	.019841	.016906	.014577	.012698	.011161
1.6		.053571	.042328	.034286	.028335	.023810	.020287	.017493	.015238	.013393
1.7		.062500	.049383	.040000	.033058	.027778	.023669	.020408	.017778	.015625
1.8		.071429	.056437	.045714	.037780	.031746	.027050	.023324	.020317	.017857
1.9		.080357	.063492	.051429	.042503	.035714	.030431	.026239	.022857	.020089
2.0		.089286	.070547	.057143	.047226	.039683	.033812	.029155	.025397	.022321
2.2		.10714	.084656	.068571	.056671	.047619	.040575	.034985	.030476	.026786
2.4		.12500	.098765	.080000	.066116	.055556	.047337	.040816	.035556	.031250
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2.8		.16071	.12698	.10286	.085006	.071429	.060862	.052478	.045714	.040179
3.0		.17857	.14109	.11429	.094451	.079365	.067625	.058309	.050794	.044643
3.2		.19653	.15520	.12571	.10390	.087302	.074387	.064440	.055873	.049107
3.4		.21429	.16931	.13714	.11334	.095238	.081150	.069971	.060952	.053571
3.6		.23214	.18342	.14857	.12279	.10317	.087912	.075802	.066032	.058036
3.8		.25000	.19753	.16000	.13223	.11111	.094674	.081633	.071111	.062500
4.0		.26786	.21164	.17143	.14168	.11905	.10144	.087464	.076190	.066964
4.2		.28571	.22575	.18286	.15112	.12698	.10820	.093294	.081270	.071429
4.4		.30357	.23986	.19429	.16057	.13492	.11496	.099125	.086349	.075893
4.6		.32143	.25397	.20571	.17001	.14286	.12172	.10494	.091429	.080357
4.8		.33929	.26808	.21714	.17946	.15079	.12849	.11075	.096508	.084821
5.0		.35714	.28219	.22857	.18890	.15873	.13525	.11656	.10159	.089286
5.2		.37500	.29630	.24000	.19835	.16667	.14201	.12237	.10667	.093750
5.4		.39286	.31041	.25143	.20779	.17460	.14877	.12818	.11175	.098214
5.6		.41071	.32451	.26286	.21724	.18254	.15554	.13399	.11683	.10268
5.8		.42857	.33862	.27429	.22668	.19048	.16230	.13980	.12190	.10714
6.0		.44643	.35273	.28571	.23613	.19841	.16906	.14561	.12698	.11161
6.2		.46429	.36684	.29714	.24557	.20635	.17582	.15142	.13206	.11607
6.4		.48214	.38095	.30857	.25502	.21429	.18259	.15723	.13714	.12054
6.6		.50000	.39506	.32000	.26446	.22222	.18935	.16305	.14222	.12500
6.8		.51786	.40907	.33143	.27391	.23016	.19611	.16886	.14730	.12946
7.0		.53571	.42308	.34286	.28335	.23810	.20287	.17467	.15238	.13393
7.2		.55357	.43709	.35429	.29280	.24603	.20964	.18048	.15746	.13839
7.4		.57143	.45110	.36571	.30224	.25397	.21640	.18629	.16254	.14286
7.6		.58929	.46511	.37714	.31169	.26190	.22316	.19210	.16762	.14732
7.8		.60714	.47912	.38857	.32113	.26984	.22992	.19791	.17270	.15178
8.0		.62500	.49313	.40000	.33058	.27778	.23669	.20372	.17778	.15625
8.2		.64286	.50714	.41143	.34002	.28571	.24345	.20953	.18286	.16071
8.4		.66071	.52115	.42286	.34947	.29365	.25021	.21534	.18794	.16518
8.6		.67857	.53515	.43429	.35891	.30159	.25697	.22115	.19302	.16964
8.8		.69643	.54916	.44571	.36836	.30952	.26374	.22697	.19810	.17411
9.0		.71429	.56317	.45714	.37780	.31746	.27050	.23278	.20317	.17857
9.2		.73214	.57718	.46857	.38725	.32540	.27726	.23859	.20825	.18304
9.4		.75000	.59119	.48000	.39669	.33333	.28402	.24440	.21333	.18750
9.6		.76786	.60520	.49142	.40614	.34127	.29079	.25021	.21841	.19196
9.8		.78571	.61921	.50284	.41558	.34921	.29755	.25602	.22349	.19643
10.0		.80357	.63322	.51426	.42503	.35714	.30431	.26183	.22857	.20089



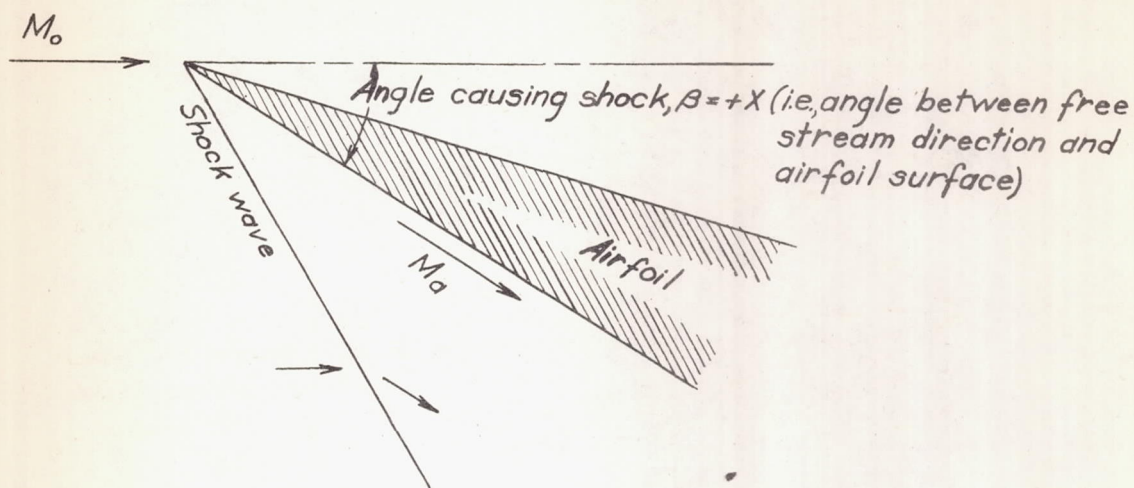
(a) Expansion at leading edge of airfoil.



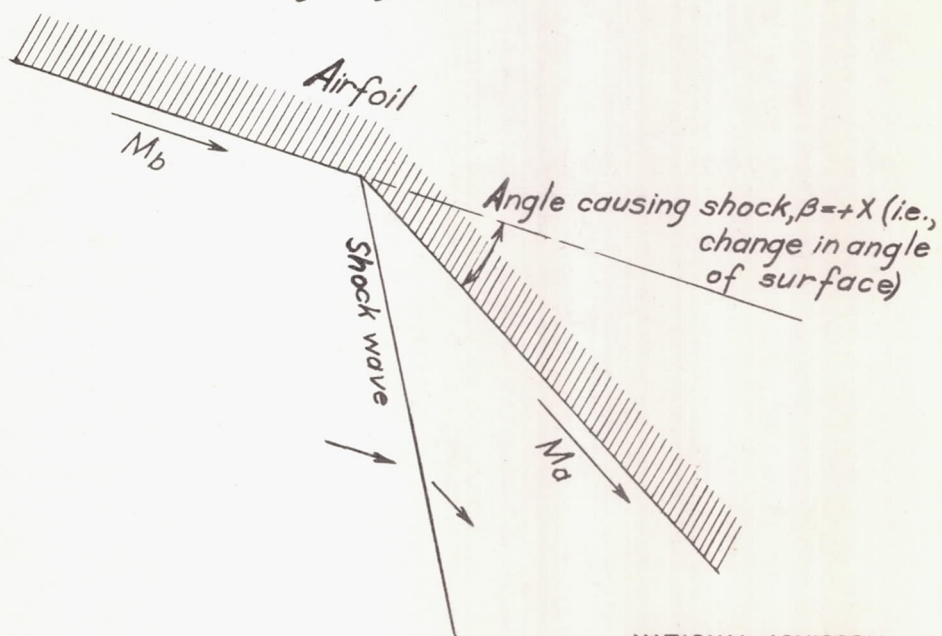
(b) Expansion along the airfoil.

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Figure 1.- Method of measuring angle causing expansion.  
The angle causing expansion is always considered negative.



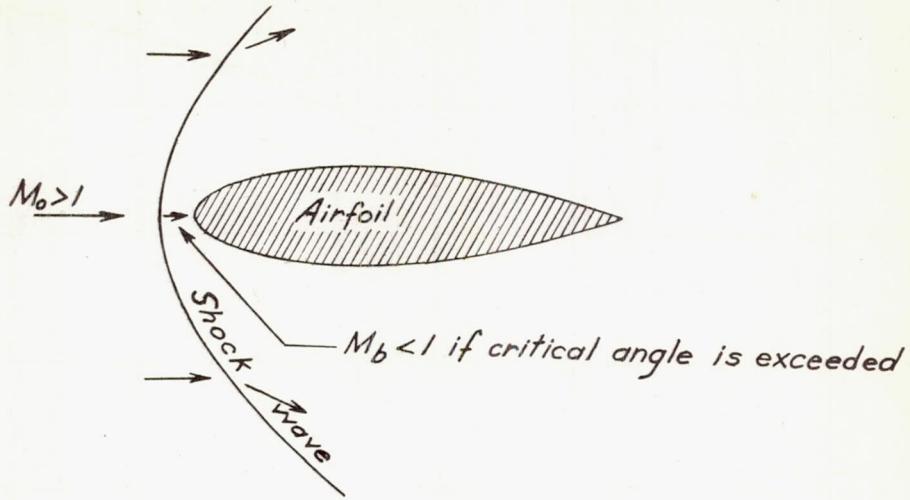
(a) Shock at leading edge of airfoil.



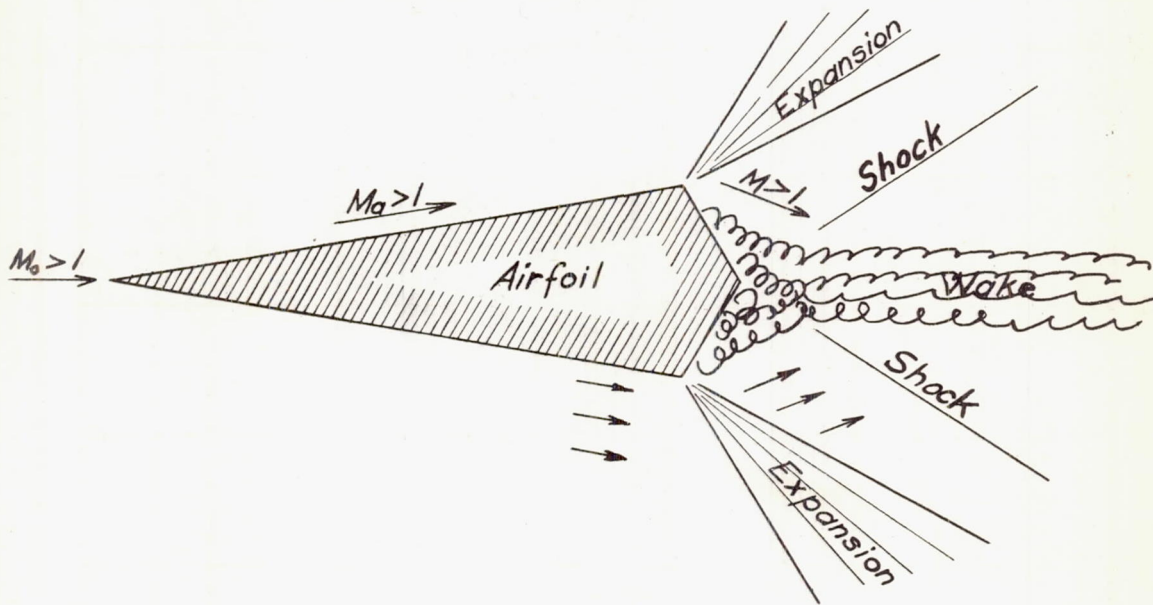
(b) Shock at intermediate point along airfoil.

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Figure 2.- Method of measuring angle causing shock. The angle causing shock is always considered positive.



(a) Shock limitation exceeded.



(b) Expansion limitations exceeded. (Turbulent wake set up)

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Figure 3.- Effect of exceeding the limitation on angles.

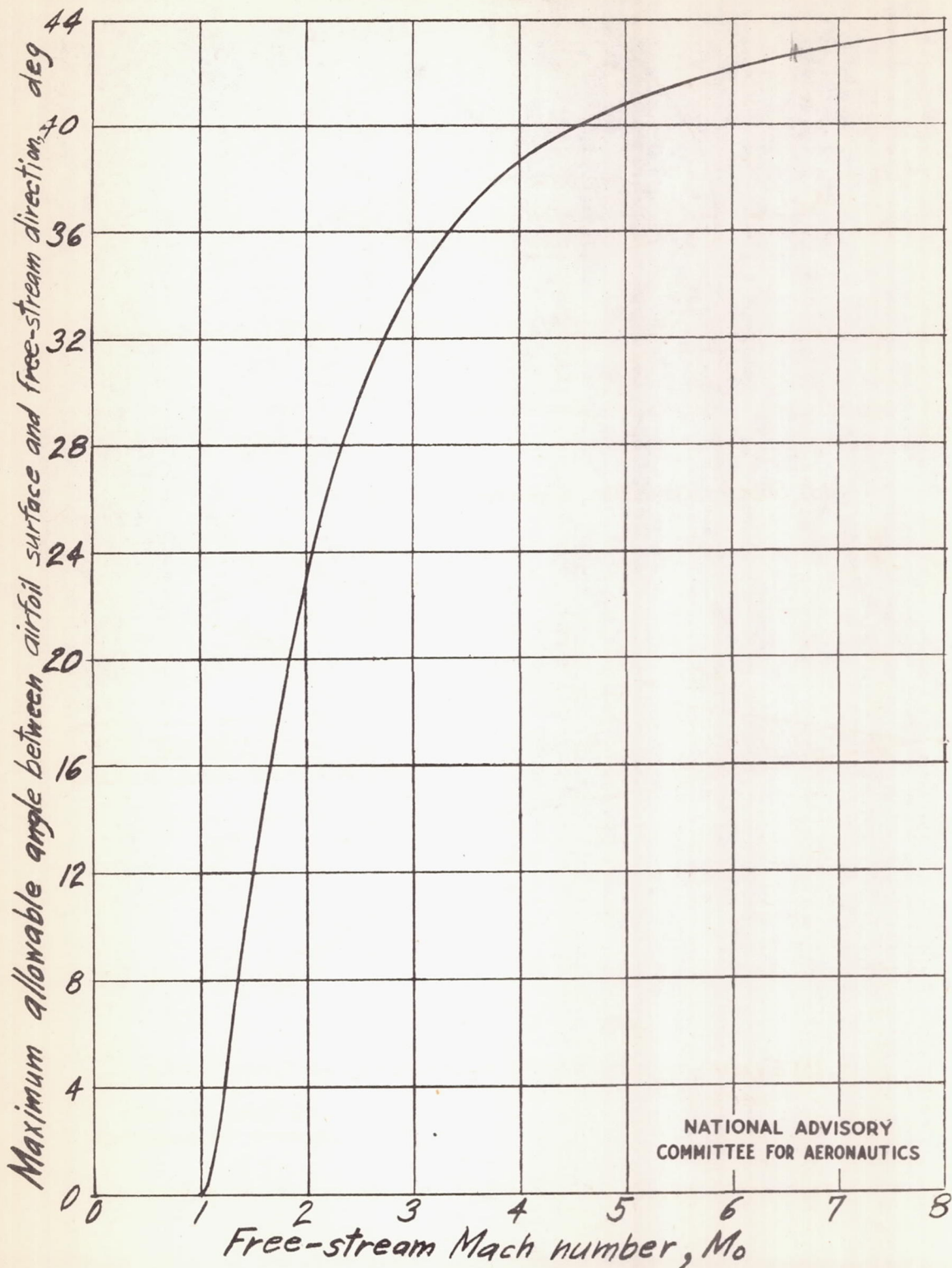


Figure 4.- Surface-angle limitation for attached shock wave.



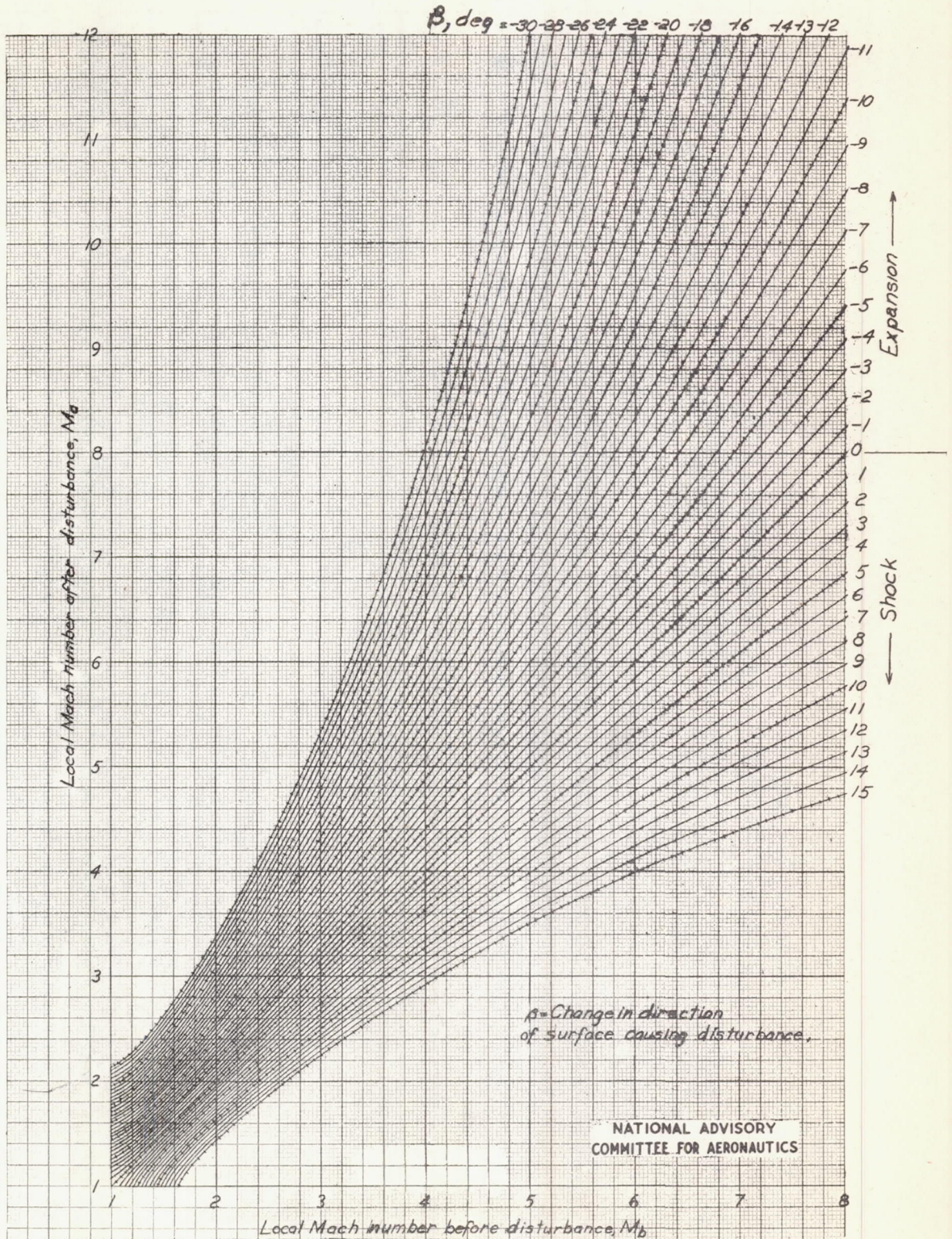


Figure 5.-Local Mach numbers before and after shocks and expansions.

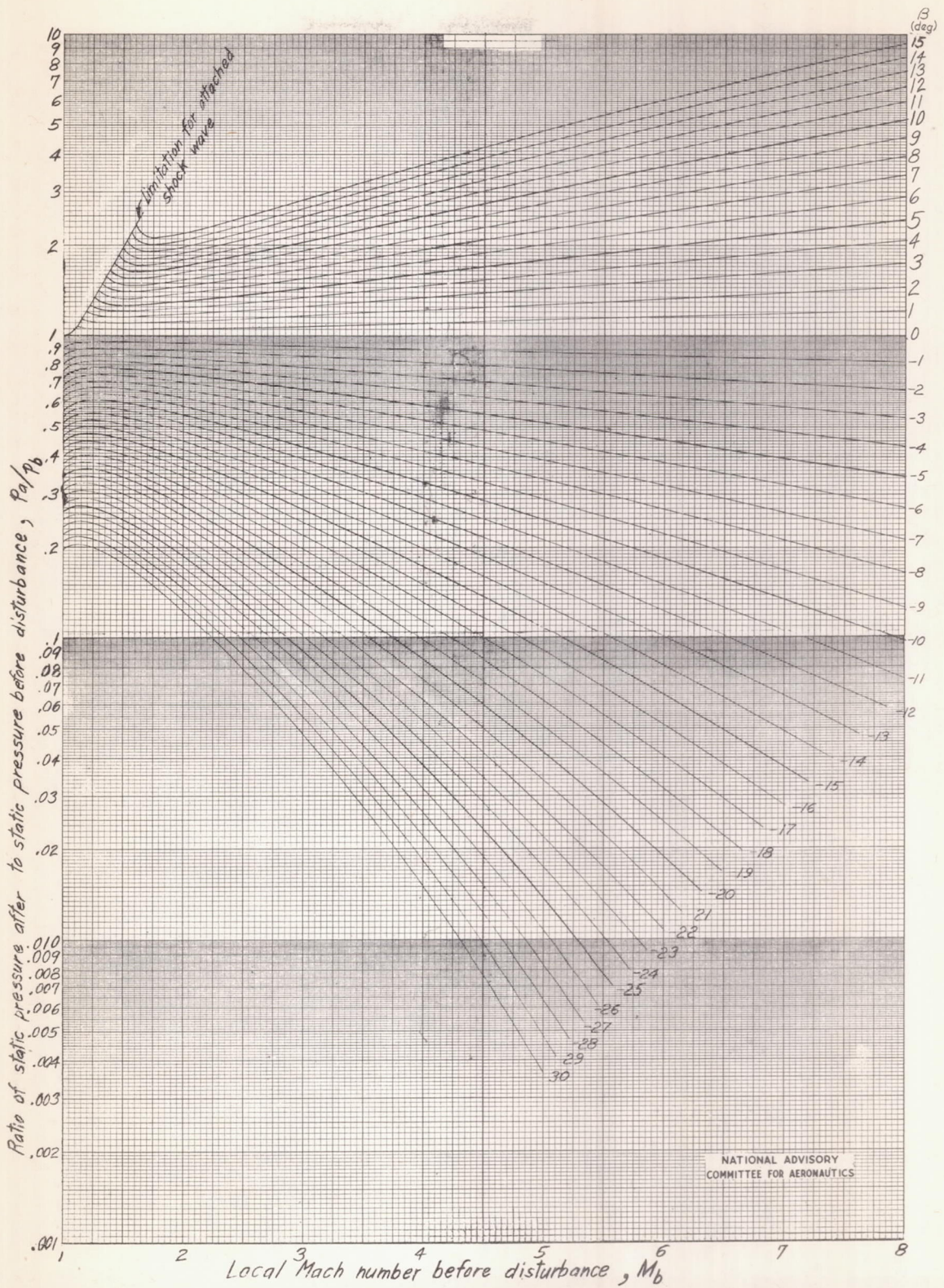
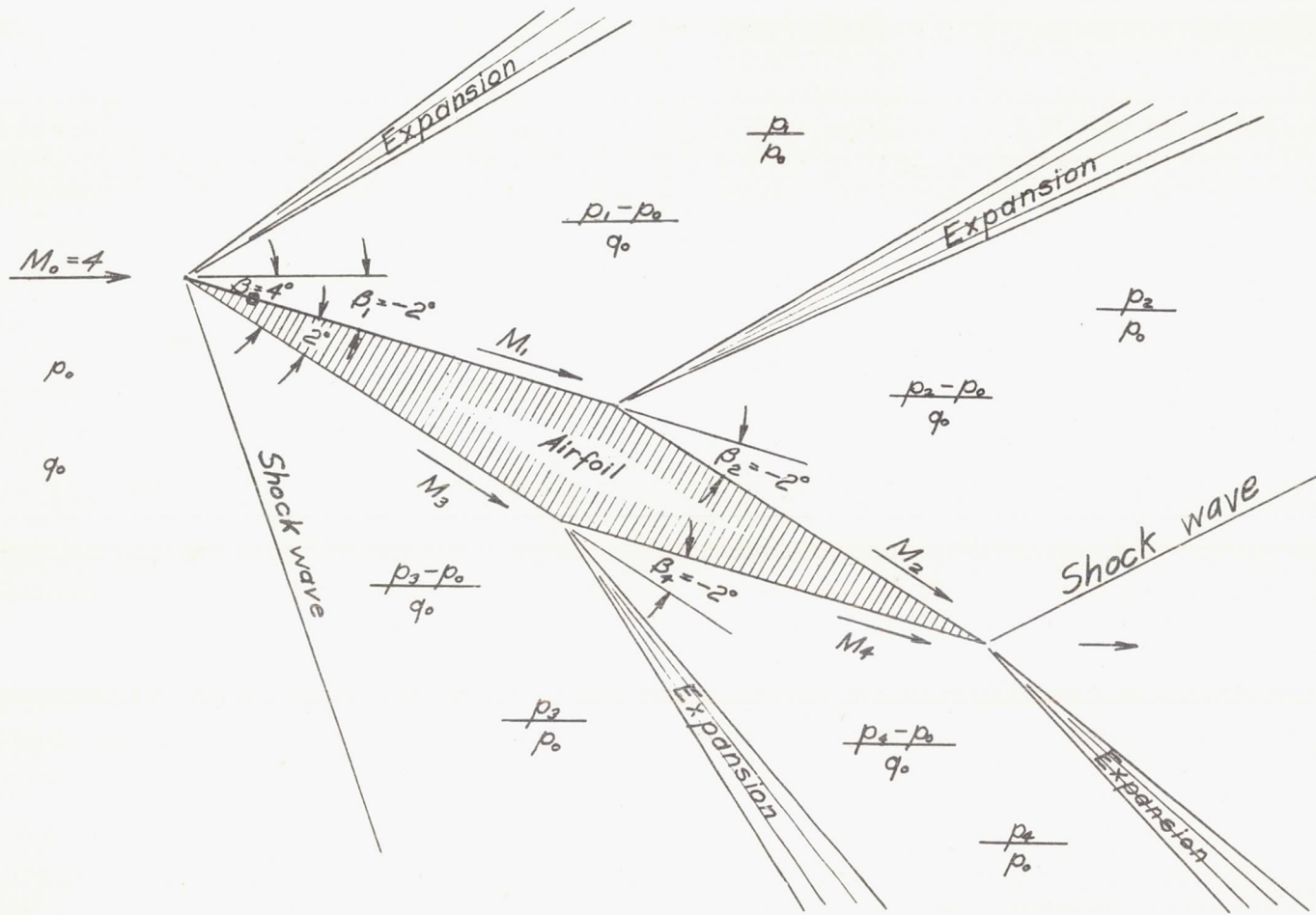


Figure 6.- Static pressure ratio across shock and expansion waves.



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Figure 7.- Example airfoil (showing conditions to be determined).

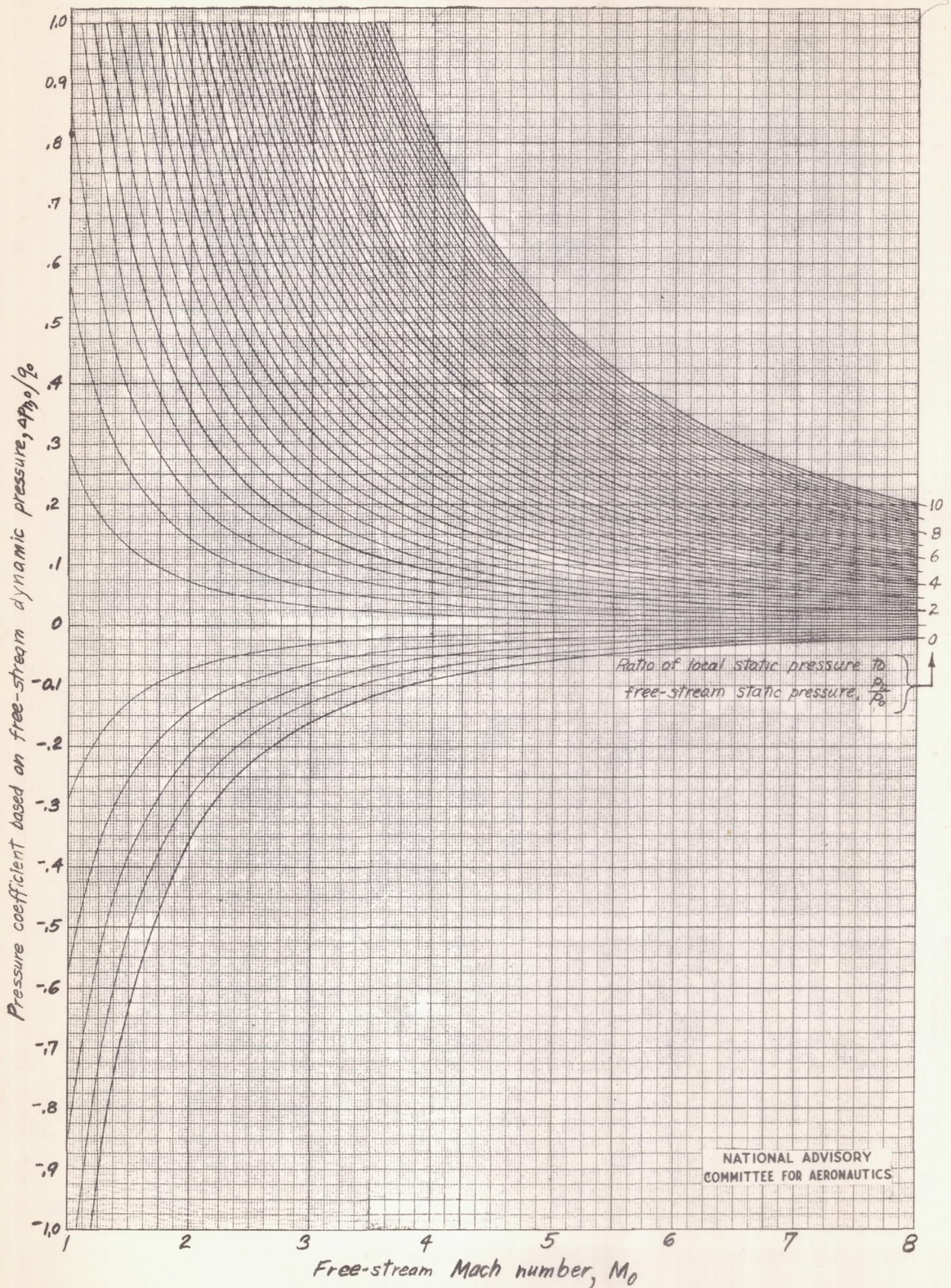


Figure 8.- Determination of pressure coefficients.

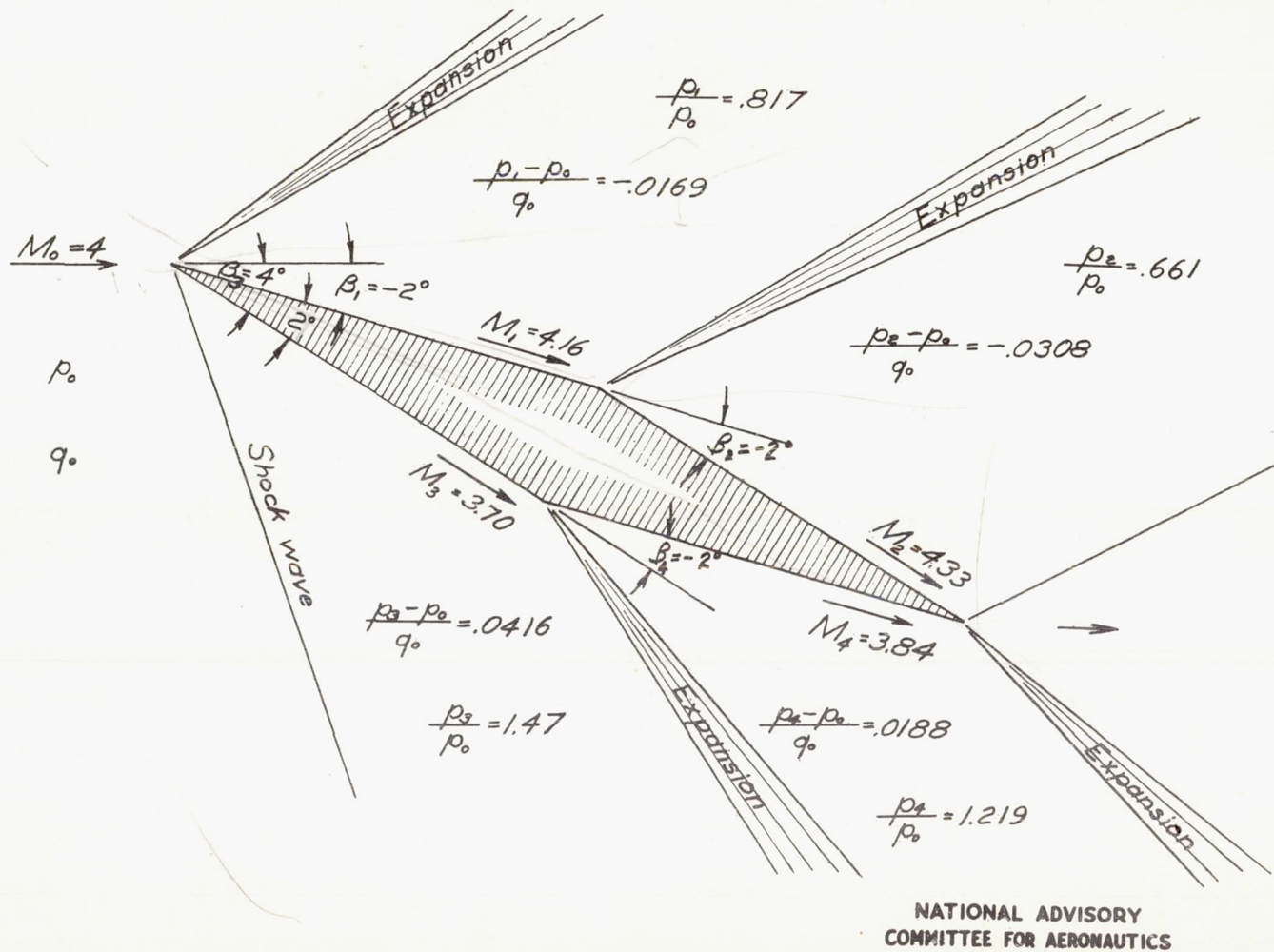
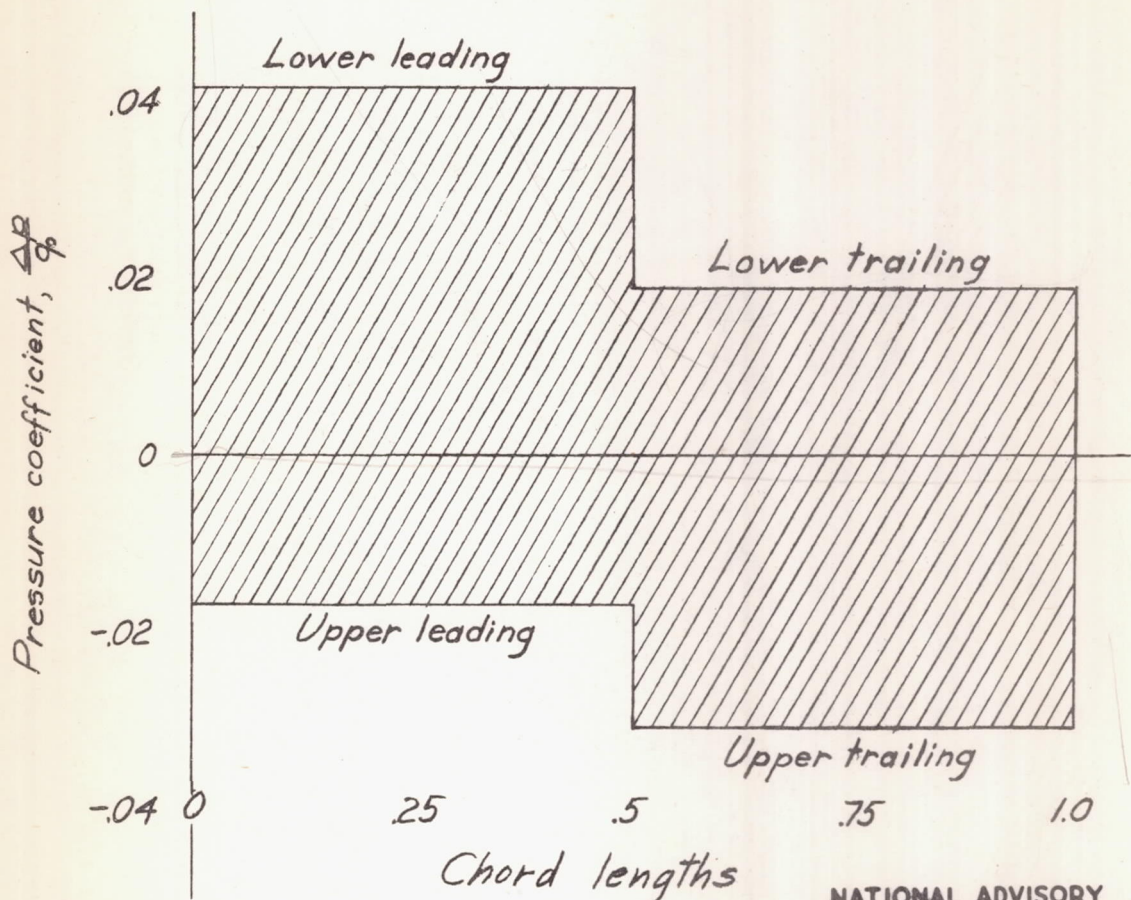
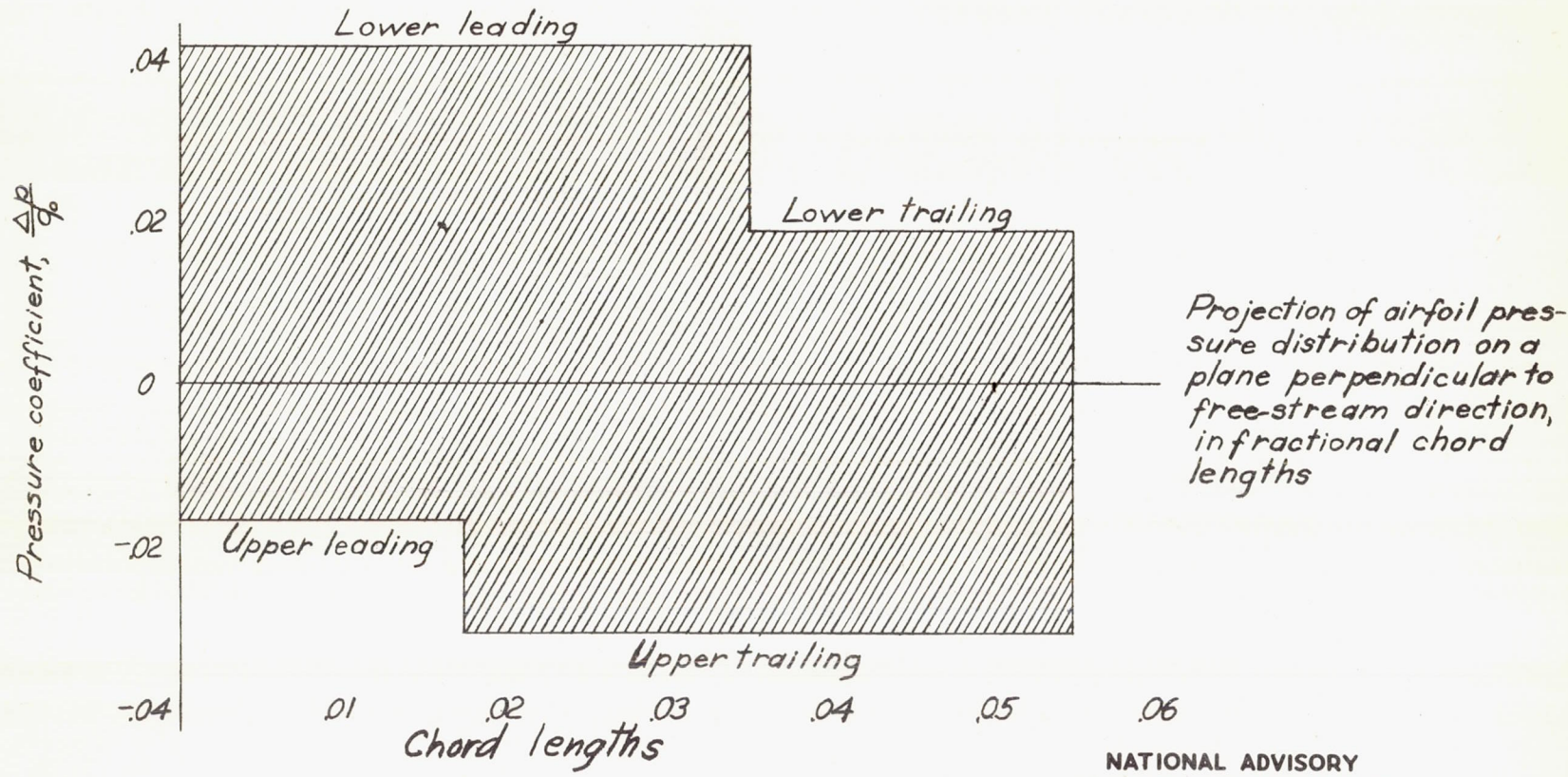


Figure 9. - Example airfoil (showing results obtained).



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Figure 10.- Determination of lift coefficient from pressure distribution for example airfoil of figures 7 and 9. Value obtained by integrating shaded area gives lift coefficient, 0.0540.



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Figure 11.- Determination of drag coefficient from pressure distribution for example airfoil of figures 7 and 9. Value obtained by integrating shaded area gives pressure drag coefficient, 0.00315.

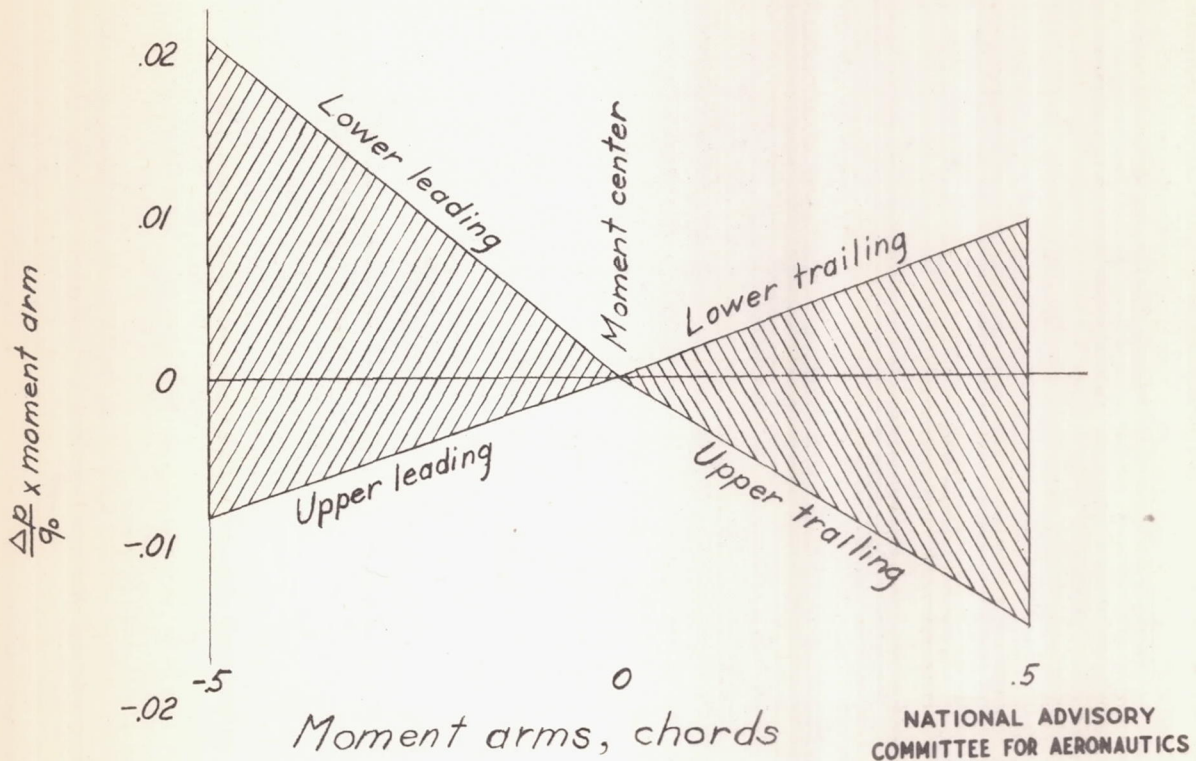


Figure 12.- Determination of moment coefficient from pressure distribution for example airfoil of figures 7 and 9. If leading and trailing surfaces give moments in the same sense add the area between "upper leading" and "lower leading" to that between "upper trailing" and "lower trailing" lines. The value obtained by integrating shaded area gives the moment coefficient about 0.50 chord, 0.0001112.



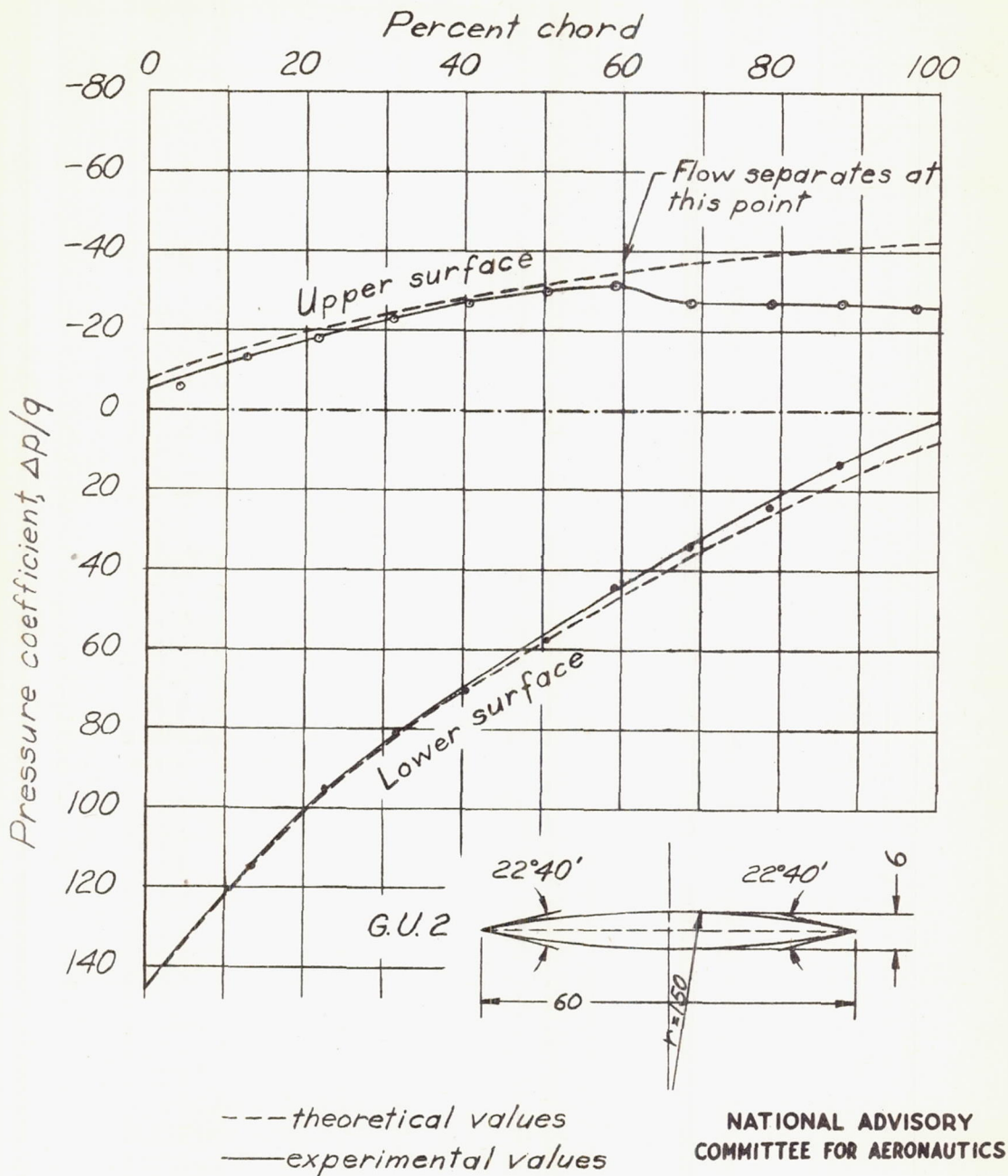


Figure 13.- Comparison of calculated and experimental results from NACA TM No. 946.  $\alpha = 14^\circ$ ;  $M_0 = 2.13$ ;  $R = 640,000$ ; thickness = 0.10 chord.

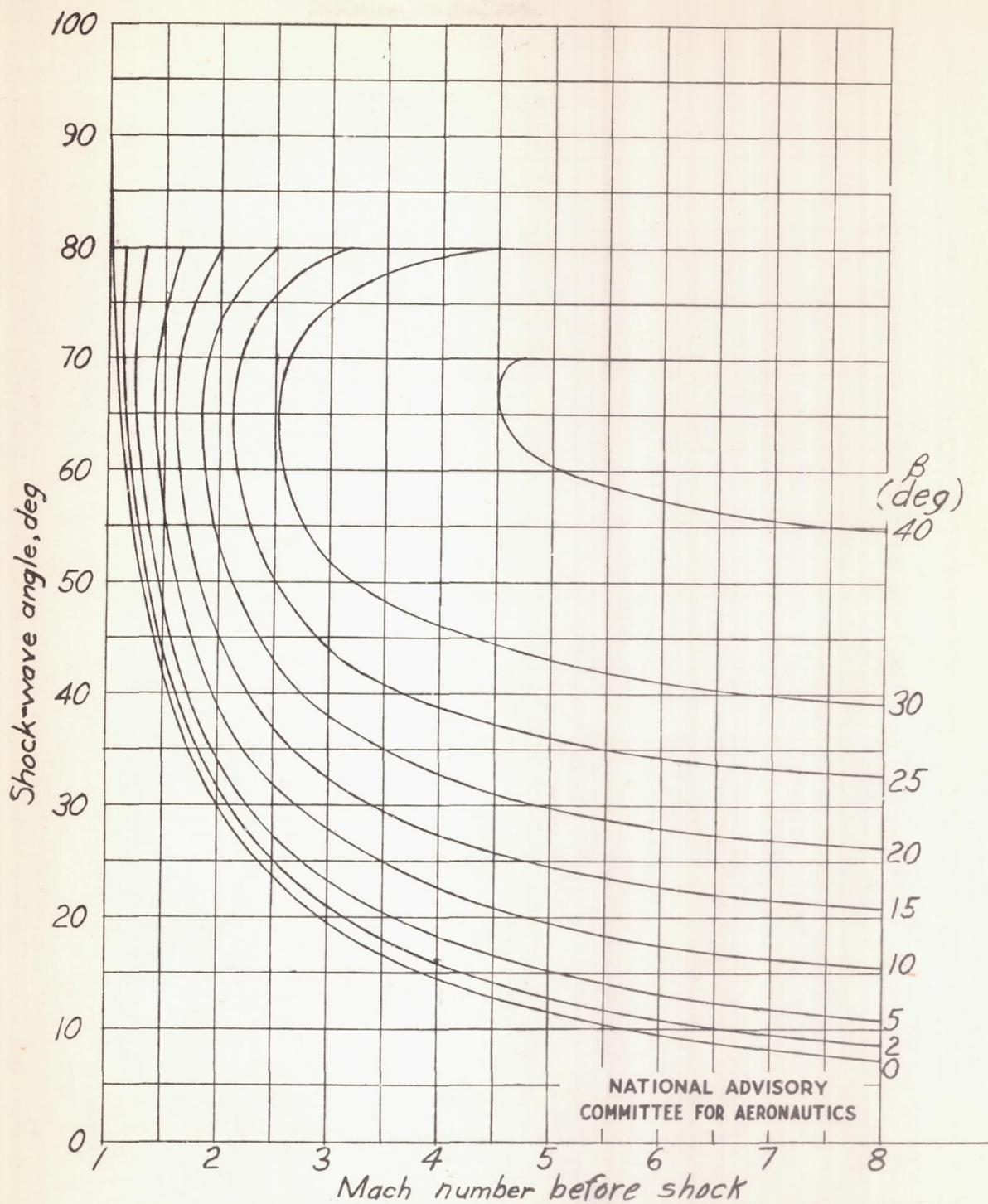


Figure 14.- Shock-wave angle for air.

