

NACA IN No. 1673

NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TECHNICAL NOTE

No. 1673

TABLES AND CHARTS OF FLOW PARAMETERS
ACROSS OBLIQUE SHOCKS

By Mary M. Neice

Langley Aeronautical Laboratory
Langley Field, Va.



Washington
August 1948

TABLES AND CHARTS OF FLOW PARAMETERS

ACROSS OBLIQUE SHOCKS

By Mary M. Neice

SUMMARY

The oblique shock-wave equations have been solved for a range of Mach number in front of the shock from 1.05 to 4.0 and for a range of shock angle from a simple Mach wave to a normal shock. The results of these calculations are presented in tabular form and include values for the Mach number in front of the shock, the shock angle, the deviation of the flow across the shock, the Mach number behind the shock, the ratio of pressure behind the shock to pressure in front of the shock, the ratio of density in front of the shock to density behind the shock, and the change in entropy across the shock. Charts of several of these parameters are also presented.

INTRODUCTION

When a compression shock occurs in supersonic flow, the flow across the shock wave undergoes changes in its physical characteristics. These changes in characteristics of the flow behind the shock have been calculated in terms of similar characteristics of the flow in front of the shock from the fundamental shock-wave equations (reference 1). Calculations of values of some of the parameters most frequently used in supersonic studies have been presented in references 2 and 3. These computations, however, are limited in scope. More recently, reference 4 has been published to give a general survey of the equations, tables, and charts that are continually being used in research on supersonic flows. The material presented herein supplements and extends the information of reference 4. Because the nature of these computations is basic and the existence of such data would serve to avoid repetition of the same calculations in the future, an attempt has been made to present the results in a form that has already proved extremely useful for rapid calculations in the study of supersonic-flow problems and in supersonic design work.

SYMBOLS

M	Mach number
p	static pressure
ρ	density
ϵ	angle of the shock
δ	angle of deviation of the flow behind the shock
β	difference between angle of shock and deviation of the flow ($\epsilon - \delta$)
ΔS	change in entropy across shock, feet ² per second ² per degree Fahrenheit
R	gas constant for air (1715 ft ² /sec ² /°F)
γ	ratio of specific heats (1.4 for air)

Subscripts

1	conditions in front of shock
2	conditions behind shock
m	maximum
s	sonic

SHOCK EQUATIONS

The flow conditions associated with the phenomena of shock in a supersonic stream are illustrated in figure 1. This diagram shows the angle of shock ϵ , the deviation of flow across the shock δ , and the angle of flow behind the shock with relation to the front of the shock β . The evaluation of the flow parameters following a shock are included herein for ready reference. The equations used for the present calculations were taken from reference 1 and are as follows:

The deviation of the flow across a shock wave is obtainable from the relation:

$$\frac{1}{\tan \delta} = \left(\frac{\gamma + 1}{2} \frac{M_1^2}{M_1^2 \sin^2 \epsilon - 1} - 1 \right) \tan \epsilon \quad (1)$$

The value of the Mach number behind the shock is determined by

$$\frac{\tan \epsilon}{\tan \beta} = \frac{2}{\gamma + 1} \left(\frac{1}{M_2^2 \sin^2 \beta} + \frac{\gamma - 1}{2} \right) \quad (2)$$

The ratio of the pressure behind the shock to the pressure in front of the shock is

$$\frac{p_2}{p_1} = \frac{2\gamma}{\gamma + 1} \left(M_1^2 \sin^2 \epsilon - \frac{\gamma - 1}{2\gamma} \right) \quad (3)$$

and the ratio of density in front of the shock to density behind the shock is

$$\frac{\rho_1}{\rho_2} = \frac{2}{\gamma + 1} \left(\frac{1}{M_1^2 \sin^2 \epsilon} + \frac{\gamma - 1}{2} \right) \quad (4)$$

The variation of entropy across the shock is given by the equation

$$\Delta S = \frac{R}{(\gamma - 1)} \left(\log_e \frac{p_2}{p_1} + \gamma \log_e \frac{\rho_1}{\rho_2} \right) \quad (5)$$

where

$$R = 1715 \text{ ft}^2/\text{sec}^2/^\circ\text{F}$$

The equation for the shock angle which gives maximum possible deviation of the flow across the shock is

$$\sin^2 \epsilon_m = \frac{1}{\gamma M_1^2} \left[\frac{\gamma + 1}{4} M_1^2 - 1 + \sqrt{(\gamma + 1) \left(1 + \frac{\gamma - 1}{2} M_1^2 + \frac{\gamma + 1}{16} M_1^4 \right)} \right] \quad (6)$$

The value of the angle of the shock which gives sonic velocity behind the shock is obtainable from the equation

$$\sin^2 \epsilon_s = \frac{1}{\gamma M_1^2} \left[\frac{\gamma + 1}{4} M_1^2 - \frac{3 - \gamma}{4} + \sqrt{(\gamma + 1) \left(\frac{9 + \gamma}{16} - \frac{3 - \gamma}{8} M_1^2 + \frac{\gamma + 1}{16} M_1^4 \right)} \right] \quad (7)$$

DISCUSSION OF TABLES AND CHARTS

The changes in Mach number, pressure, density, entropy, and the deviation of the flow across the shock wave are presented in tabular form and on the charts. Table I includes solutions of shock-wave equations for stream Mach numbers from $M_1 = 1.05$ to $M_1 = 4.0$ and for a range of angles of shock from a simple Mach wave to an angle normal to the stream (normal shock). For each angle of shock, the ratio of the pressure behind the shock to the pressure in front of the shock, the ratio of the density in front of the shock to the density behind the shock, the deviation of the flow across the shock, and the change in entropy across the shock are given. Solutions of the shock-wave equations which give Mach numbers behind the shock of less than one are included because they are useful in the study of detached shock, internal flows, and similar fields. Table II gives values of the angle of the shock and the Mach number behind the shock for the maximum possible deviation of the flow as well as values of the angle of the shock and the deviation of the flow which gives sonic velocity behind the shock ($M_2 = 1.0$).

The data given in table I were plotted and from these curves values of the angle of the shock and the Mach number behind the shock were taken for even values of the deviation of the flow at Mach numbers in front of the shock from 1.05 to 4.0. These values (read from faired curves) are given in table III and are presented solely to aid in the preparation of large-scale figures similar to figure 2, which shows the variation of the angle of the shock with the Mach number in front of the shock for values of the deviation of the flow from zero to the maximum deviation. In figure 3 the variation of the Mach number behind the shock with the Mach number in front of the shock is shown for the same values of the deviation of the flow. Figure 4 gives the variation of the angle of shock with the pressure ratio p_2/p_1 for a range of

Mach number in front of the shock from $M_1 \cong 1.05$ to $M_1 = 4.0$. The variation of the maximum deviation of the flow across the shock with the Mach number in front of the shock is shown in figure 5.

Langley Aeronautical Laboratory
National Advisory Committee for Aeronautics
Langley Field, Va., March 31, 1948

REFERENCES

1. Crocco, Luigi: Singolarità della corrente gassosa iperacustica nell'intorno di una prora a diedro. *L'Aerotecnica*, vol. XVII, fasc. 6, June 1937, pp. 519-534.
2. Ivey, H. Reese, Stickle, George W., and Schuettler, Alberta: Charts for Determining the Characteristics of Sharp-Nose Airfoils in Two-Dimensional Flow at Supersonic Speeds. NACA TN No. 1143, 1947.
3. Moeckel, W. E., and Connors, J. F.: Charts for the Determination of Supersonic Air Flow against Inclined Planes and Axially Symmetric Cones. NACA TN No. 1373, 1947.
4. The Staff of the Ames 1- by 3-Foot Supersonic Wind-Tunnel Section: Notes and Tables for Use in the Analysis of Supersonic Flow. NACA TN No. 1428, 1947.

TABLE I - Concluded

VALUES OF DEVIATION OF FLOW, MACH NUMBER, PRESSURE RATIO, DENSITY RATIO,
AND CHANGE IN ENTROPY ACROSS SHOCK WAVES - Concluded

Table with multiple columns: M1 (deg, min), delta (deg, min), M2, P2/P1, rho1/rho2, delta S (ft^2/lb sec^2), and repeated columns for M1, delta, M2, P2/P1, rho1/rho2, delta S. Rows correspond to Mach numbers 3.70, 3.75, 3.80, 3.85, and 4.00.

TABLE II

ANGLE OF SHOCK AND M_{2m} FOR MAXIMUM DEVIATION OF THE FLOW
 AND ANGLE OF SHOCK AND δ , WHICH GIVE
 SONIC VELOCITY BEHIND THE SHOCK

M_1	ϵ_m		ϵ_s		δ_m		δ_s		M_{2m}
	(deg)	(min)	(deg)	(min)	(deg)	(min)	(deg)	(min)	
1.0	90		90		0		0		1.000
1.1	76	18	73	14	1	31	1	24	.9710
1.2	71	59	68	5	3	57	3	42	.9500
1.3	69	24	65	7	6	40	6	19	.9357
1.4	67	42	63	20	9	26	9	1	.9268
1.5	66	36	62	15	12	6	11	41	.9212
1.6	65	50	61	39	14	39	14	15	.9187
1.7	65	19	61	22	17	0	16	38	.9185
1.8	64	59	61	17	19	11	18	50	.9196
1.9	64	47	61	21	21	10	20	52	.9216
2.0	64	40	61	29	22	59	22	43	.9243
2.1	64	37	61	41	24	37	24	23	.9274
2.2	64	37	61	54	26	6	25	54	.9306
2.3	64	41	62	9	27	28	27	17	.9331
2.4	64	42	62	24	28	42	28	32	.9374
2.5	64	48	62	39	29	48	29	40	.9397
2.6	64	52	62	53	30	49	30	42	.9426
2.7	64	57	63	7	31	45	31	39	.9464
2.8	65	3	63	21	32	35	32	30	.9489
2.9	65	9	63	33	33	21	33	17	.9514
3.0	65	15	63	46	34	4	34	1	.9537
3.1	65	20	63	58	34	44	34	40	.9565
3.2	65	25	64	8	35	20	35	17	.9589
3.3	65	31	64	18	35	53	35	51	.9606
3.4	65	36	64	28	36	24	36	22	.9627
3.5	65	41	64	37	36	52	36	50	.9645
3.6	65	46	64	45	37	18	37	17	.9660
3.7	65	51	64	54	37	43	37	41	.9674
3.8	65	56	65	2	38	5	38	4	.9685
3.9	65	59	65	9	38	27	38	25	.9708
4.0	66	3	65	15	38	47	38	45	.9721

TABLE III
VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK

M ₁	δ		ε		M ₂	M ₁	δ		ε		M ₂		
	(deg)	(min)	(deg)	(min)			(deg)	(min)	(deg)	(min)			
1.05	0	0	72	15	1.050	1.40	5	0	52	46	1.216		
	0	30	78	0	1.000		5	30	53	39	1.195		
1.10	0	0	65	23	1.100	1.45	6	0	54	38	1.174		
	0	30	67	20	1.074		6	30	55	40	1.151		
	1	0	69	54	1.041		7	0	56	46	1.127		
	1	30	74	40	.982		7	30	57	58	1.102		
1.15	0	0	60	24	1.150	1.50	8	0	59	22	1.074		
	0	30	61	42	1.127		8	30	61	6	1.042		
	1	0	63	11	1.104		9	0	63	14	1.004		
	1	30	64	54	1.075		0	0	43	36	1.450		
	2	0	66	58	1.044		0	30	44	11	1.433		
	2	30	70	38	.998		1	0	44	48	1.416		
1.20	0	0	56	26	1.200	1.55	1	30	45	24	1.397		
	0	30	57	27	1.180		2	0	46	2	1.380		
	1	0	58	54	1.158		2	30	46	40	1.362		
	1	30	59	45	1.135		3	0	47	10	1.344		
	2	0	61	3	1.110		3	30	48	1	1.326		
	2	30	62	32	1.083		4	0	48	42	1.307		
	3	0	64	25	1.053		4	30	49	24	1.290		
	3	30	66	56	1.019		5	0	50	7	1.271		
	1.25	0	0	53	8		1.250	1.60	5	30	50	55	1.252
		0	30	53	58		1.232		6	0	51	45	1.232
1		0	54	54	1.212	6	30		52	36	1.211		
1		30	55	51	1.192	7	0		53	30	1.190		
2		0	56	52	1.171	7	30		54	28	1.168		
2		30	57	58	1.149	8	0		55	30	1.146		
3		0	59	10	1.126	8	30		56	38	1.120		
3		30	60	29	1.100	9	0		57	54	1.095		
4		0	62	5	1.072	9	30		59	19	1.067		
4		30	63	59	1.040	10	0		61	5	1.036		
5		0	66	32	.998	10	30		63	38	.987		
1.30		0	0	50	17	1.300	1.65		0	0	41	49	1.500
	0	30	51	2	1.282	0		30	42	22	1.487		
	1	0	51	50	1.264	1		0	42	55	1.466		
	1	30	52	39	1.245	1		30	43	30	1.449		
	2	0	53	30	1.225	2		0	44	5	1.431		
	2	30	54	24	1.206	2		30	44	42	1.414		
	3	0	55	11	1.186	3		0	45	19	1.396		
	3	30	56	20	1.165	3		30	45	55	1.380		
	4	0	57	26	1.142	4		0	46	34	1.362		
	4	30	58	36	1.117	4		30	47	14	1.344		
	5	0	59	58	1.091	5		0	47	54	1.326		
	5	30	61	32	1.062	5		30	48	37	1.309		
	6	0	63	39	1.028	6		0	49	22	1.290		
	6	30	66	28	.980	6		30	50	6	1.271		
1.35	0	0	47	47	1.350	1.70	7	0	50	54	1.251		
	0	30	48	29	1.332		7	30	51	44	1.230		
	1	0	49	10	1.315		8	0	52	34	1.209		
	1	30	49	55	1.297		8	30	53	30	1.187		
	2	0	50	41	1.278		9	0	54	26	1.164		
	2	30	51	18	1.259		9	30	55	30	1.139		
	3	0	52	16	1.240		10	0	56	42	1.113		
	3	30	53	6	1.219		10	30	57	58	1.086		
	4	0	53	59	1.199		11	0	59	29	1.055		
	4	30	54	56	1.177		11	30	61	25	1.016		
	5	0	55	57	1.156		12	0	64	24	.956		
	5	30	57	3	1.131		1.75	0	0	40	11	1.550	
	6	0	58	17	1.106			0	30	40	41	1.532	
	6	30	59	37	1.080			1	0	41	14	1.515	
	7	0	61	14	1.050			1	30	41	47	1.497	
	7	30	63	15	1.017			2	0	42	10	1.480	
8	0	66	58	.952	2	30		42	54	1.463			
1.40	0	0	45	35	1.400	3		0	43	29	1.446		
	0	30	46	10	1.382	3		30	44	4	1.428		
	1	0	46	49	1.365	4		0	44	41	1.412		
	1	30	47	28	1.347	4		30	45	18	1.395		
	2	0	48	9	1.330	5		0	45	55	1.376		
	2	30	48	52	1.311	5		30	46	34	1.360		
	3	0	49	35	1.293	6		0	47	14	1.341		
	3	30	50	19	1.275	6		30	47	55	1.324		
	4	0	51	6	1.256	7		0	48	40	1.306		
	4	30	51	55	1.236	7		30	49	24	1.286		
						8	0	50	10	1.267			
						8	30	50	59	1.246			
					9	0	51	50	1.225				
					9	30	52	41	1.204				



TABLE III - Continued

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Continued

M ₁	δ		ε		M ₂	M ₁	δ		ε		M ₂
	(deg)	(min)	(deg)	(min)			(deg)	(min)	(deg)	(min)	
1.55	10	0	53	17	1.181	1.70	3	30	39	26	1.580
	10	30	54	36	1.156		4	0	39	58	1.563
	11	0	55	38	1.131		4	30	40	30	1.546
	11	30	56	52	1.105		5	0	41	2	1.528
	12	0	58	14	1.075		5	30	41	35	1.511
	12	30	59	49	1.040		6	0	42	9	1.494
	13	0	62	0	1.000		6	30	42	43	1.476
1.60	0	0	38	41	1.600	7	0	43	18	1.458	
	0	30	39	14	1.582	7	30	43	54	1.440	
	1	0	39	47	1.565	8	0	44	30	1.421	
	1	30	40	19	1.548	8	30	45	10	1.403	
	2	0	40	53	1.531	9	0	45	49	1.379	
	2	30	41	27	1.515	9	30	46	29	1.365	
	3	0	42	1	1.497	10	0	47	10	1.346	
	3	30	42	34	1.480	10	30	47	53	1.327	
	4	0	43	10	1.462	11	0	48	37	1.307	
	4	30	43	45	1.445	11	30	49	22	1.287	
	5	0	44	19	1.427	12	0	50	9	1.267	
	5	30	44	55	1.410	12	30	51	0	1.245	
	6	0	45	32	1.391	13	0	51	53	1.223	
	6	30	46	10	1.374	13	30	52	47	1.200	
	7	0	46	47	1.356	14	0	53	48	1.175	
	7	30	47	26	1.338	14	30	54	52	1.149	
	8	0	48	7	1.320	15	0	56	0	1.121	
	8	30	48	50	1.300	15	30	57	18	1.091	
	9	0	49	32	1.281	16	0	58	42	1.057	
	9	30	50	14	1.261	16	30	60	40	1.014	
	10	0	51	6	1.240	17	0	64	40	.930	
10	30	51	58	1.218	1.75	0	0	34	51	1.750	
11	0	52	53	1.196		0	30	35	18	1.732	
11	30	53	53	1.174		1	0	35	47	1.716	
12	0	54	56	1.148		1	30	36	15	1.698	
12	30	56	1	1.122		2	0	36	44	1.681	
13	0	57	14	1.093		2	30	37	13	1.664	
13	30	58	44	1.062		3	0	37	42	1.647	
14	0	60	37	1.024		3	30	38	11	1.630	
14	30	63	20	.965		4	0	38	42	1.612	
1.65	0	0	37	18		1.650	4	30	39	11	1.595
	0	30	37	47		1.632	5	0	39	42	1.578
	1	0	38	16		1.615	5	30	40	14	1.561
	1	30	38	45		1.598	6	0	40	47	1.544
	2	0	39	15		1.581	6	30	41	20	1.526
	2	30	39	46	1.564	7	0	41	54	1.509	
	3	0	40	16	1.552	7	30	42	28	1.490	
	3	30	40	49	1.530	8	0	43	2	1.471	
	4	0	41	22	1.514	8	30	43	38	1.452	
	4	30	41	56	1.496	9	0	44	14	1.435	
	5	0	42	31	1.479	9	30	44	32	1.416	
	5	30	43	5	1.461	10	0	45	29	1.397	
	6	0	43	41	1.444	10	30	46	10	1.379	
	6	30	44	17	1.425	11	0	46	50	1.360	
	7	0	44	55	1.408	11	30	47	32	1.341	
	7	30	45	32	1.389	12	0	48	18	1.322	
	8	0	46	11	1.370	12	30	49	4	1.300	
	8	30	46	52	1.351	13	0	49	51	1.280	
	9	0	47	33	1.332	13	30	50	39	1.257	
	9	30	47	15	1.313	14	0	51	32	1.234	
	10	0	49	1	1.294	14	30	52	28	1.211	
10	30	49	47	1.274	15	0	53	26	1.186		
11	0	50	36	1.252	15	30	54	28	1.160		
11	30	51	26	1.232	16	0	55	38	1.131		
12	0	52	21	1.210	16	30	56	54	1.101		
12	30	53	16	1.187	17	0	58	22	1.067		
13	0	54	16	1.163	17	30	60	8	1.027		
13	30	55	22	1.136	18	0	63	6	.963		
14	0	56	33	1.108	1.80	0	0	33	45	1.800	
14	30	57	54	1.077		0	30	34	13	1.785	
15	0	59	30	1.043		1	0	34	39	1.767	
15	30	61	44	.997		1	30	35	6	1.750	
1.70	0	0	36	2		1.700	2	0	35	34	1.727
	0	30	36	30		1.684	2	30	36	2	1.714
	1	0	36	58		1.667	3	0	36	30	1.696
	1	30	37	28	1.649	3	30	36	58	1.678	
	2	0	37	56	1.632	4	0	37	48	1.661	
	2	30	38	26	1.615	4	30	37	59	1.644	
	3	0	38	55	1.597	5	0	38	9	1.626	
					5	30	39	0	1.609		
					6	0	39	30	1.592		



TABLE III - Continued

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Continued

M ₁	δ		ε		M ₂	M ₁	δ		ε		M ₂	
	(deg)	(min)	(deg)	(min)			(deg)	(min)	(deg)	(min)		
1.80	6	30	40	2	1.575	1.90	5	0	36	13	1.725	
	7	0	40	34	1.557		5	30	36	42	1.707	
	7	30	41	8	1.540		6	0	37	13	1.690	
	8	0	41	42	1.523		6	30	37	43	1.672	
	8	30	42	16	1.505		7	0	38	14	1.654	
	9	0	42	50	1.487		7	30	38	45	1.636	
	9	30	43	26	1.468		8	0	39	17	1.618	
	10	0	44	2	1.450		8	30	39	49	1.600	
	10	30	44	42	1.431		9	0	40	21	1.582	
	11	0	45	22	1.411		9	30	40	54	1.565	
	11	30	46	1	1.392		10	0	41	29	1.547	
	12	0	46	41	1.372		10	30	42	4	1.528	
	12	30	47	24	1.352		11	0	42	40	1.510	
	13	0	48	9	1.332		11	30	43	16	1.490	
	13	30	48	54	1.311		12	0	43	52	1.471	
	14	0	49	39	1.289		12	30	44	30	1.452	
	14	30	50	28	1.267		13	0	45	10	1.433	
	15	0	51	20	1.243		13	30	45	51	1.413	
	15	30	52	14	1.219		14	0	46	33	1.392	
16	0	53	10	1.195	14	30	47	16	1.371			
16	30	54	15	1.170	15	0	48	0	1.349			
17	0	55	22	1.142	15	30	48	44	1.327			
17	30	56	36	1.112	16	0	49	32	1.305			
18	0	58	0	1.077	16	30	50	20	1.272			
18	30	59	42	1.035	17	0	51	12	1.259			
19	0	62	18	.972	17	30	52	4	1.234			
1.85	0	0	32	43	1.850	1.95	18	0	53	2	1.209	
	0	30	33	9	1.832		18	30	54	5	1.182	
	1	0	33	35	1.816		19	0	55	14	1.152	
	1	30	34	2	1.797		19	30	56	29	1.120	
	2	0	34	29	1.780		20	0	57	52	1.084	
	2	30	34	56	1.764		20	30	59	36	1.043	
	3	0	35	23	1.746		21	0	62	6	.981	
	3	30	35	50	1.729		1.95	0	0	30	51	1.950
	4	0	36	19	1.711			0	30	31	16	1.932
	4	30	36	47	1.694			1	0	31	42	1.915
	5	0	37	17	1.676			1	30	32	7	1.897
	5	30	37	47	1.659			2	0	32	33	1.879
	6	0	38	18	1.641			2	30	32	59	1.862
	6	30	38	49	1.624			3	0	33	25	1.844
	7	0	39	20	1.607			3	30	33	51	1.826
	7	30	40	12	1.590			4	0	34	19	1.809
	8	0	40	25	1.571			4	30	34	46	1.791
	8	30	40	58	1.553			5	0	35	14	1.773
	9	0	41	34	1.535			5	30	35	42	1.756
	9	30	42	9	1.516			6	0	36	11	1.738
10	0	42	43	1.498	6	30		36	41	1.720		
10	30	43	19	1.480	7	0		37	10	1.703		
11	0	43	56	1.460	7	30		37	42	1.685		
11	30	44	33	1.441	8	0		38	12	1.666		
12	0	45	12	1.423	8	30		38	44	1.647		
12	30	45	53	1.403	9	0		39	15	1.629		
13	0	46	14	1.384	9	30		39	48	1.611		
13	30	47	18	1.364	10	0	40	20	1.593			
14	0	48	2	1.344	10	30	40	54	1.575			
14	30	48	46	1.322	11	0	41	29	1.556			
15	0	49	33	1.300	11	30	42	5	1.537			
15	30	50	22	1.277	12	0	42	40	1.519			
16	0	51	13	1.254	12	30	43	16	1.500			
16	30	52	8	1.230	13	0	43	55	1.480			
17	0	53	6	1.204	13	30	44	34	1.459			
17	30	54	6	1.177	14	0	45	14	1.439			
18	0	55	14	1.147	14	30	45	55	1.418			
18	30	56	26	1.117	15	0	46	36	1.397			
19	0	57	48	1.083	15	30	47	20	1.376			
19	30	59	28	1.042	16	0	48	4	1.355			
20	0	62	14	.983	16	30	48	50	1.333			
1.90	0	0	31	45	1.900	17	0	49	37	1.310		
	0	30	32	9	1.884	17	30	50	26	1.286		
	1	0	32	35	1.866	18	0	51	19	1.261		
	1	30	33	1	1.847	18	30	52	14	1.236		
	2	0	33	28	1.830	19	0	53	10	1.210		
	2	30	33	54	1.813	19	30	54	15	1.182		
	3	0	34	22	1.795	20	0	55	22	1.152		
	3	30	34	48	1.777	20	30	56	38	1.121		
	4	0	35	16	1.760	21	0	58	6	1.084		
	4	30	35	44	1.743	21	30	59	46	1.038		
					22	0	62	10	.963			



TABLE III - Continued

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Continued

M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂	
		(deg)	(min)				(deg)	(min)				(deg)	(min)		
2.00	0	30	0	2.000	2.10	18	47	13	1.407	2.25	4	29	32	2.096	
	1	30	50	1.964		19	48	44	1.361		5	30	24	2.059	
	2	31	40	1.927		20	50	22	1.312		6	31	16	2.021	
	3	32	30	1.892		21	52	7	1.260		7	32	10	1.982	
	4	33	23	1.856		22	54	9	1.202		8	33	5	1.944	
	5	34	19	1.821		23	56	34	1.136		9	34	2	1.905	
	6	35	16	1.785		24	59	48	1.051		10	35	2	1.866	
	7	36	13	1.749		2.15	0	27	43		2.150	11	36	1	1.827
	8	37	14	1.713			1	28	30		2.112	12	37	4	1.788
	9	38	16	1.676			2	29	19		2.075	13	38	8	1.747
	10	39	19	1.639			3	30	7		2.039	14	39	14	1.706
	11	40	25	1.602			4	30	57		2.002	15	40	24	1.666
	12	41	33	1.565			5	31	48		1.964	16	41	36	1.624
	13	42	44	1.526			6	32	42		1.927	17	42	50	1.581
	14	44	1	1.486			7	33	37		1.890	18	44	8	1.538
	15	45	20	1.447			8	34	34		1.852	19	45	29	1.495
	16	46	42	1.405			9	35	33		1.814	20	46	56	1.447
	17	48	10	1.360		10	36	34	1.776		21	48	27	1.399	
	18	49	42	1.313		11	37	38	1.739		22	50	4	1.347	
	19	51	9	1.264		12	38	43	1.700		23	51	50	1.292	
	20	53	20	1.211		13	39	49	1.662		24	53	49	1.233	
	21	55	32	1.151		14	40	58	1.620		25	56	8	1.164	
22	58	24	1.077	15	42	10	1.579	26	59	8	1.080				
2.05	0	29	12	2.050	2.20	16	43	25	1.537	2.30	0	25	46	2.300	
	1	30	0	2.014		17	44	43	1.495		1	26	30	2.260	
	2	30	50	1.978		18	46	5	1.451		2	27	16	2.222	
	3	31	40	1.943		19	47	33	1.406		3	28	4	2.183	
	4	32	32	1.907		20	49	8	1.358		4	28	54	2.144	
	5	33	23	1.871		21	50	48	1.307		5	29	44	2.105	
	6	34	22	1.834		22	52	36	1.252		6	30	35	2.067	
	7	35	18	1.797		23	54	42	1.192		7	31	30	2.028	
	8	36	16	1.760		24	57	14	1.122		8	32	25	1.989	
	9	37	18	1.723		25	60	52	1.033		9	33	21	1.949	
	10	38	22	1.685		2.25	0	27	2		2.200	10	34	19	1.910
	11	39	26	1.648			1	27	48		2.161	11	35	20	1.870
	12	40	32	1.609			2	28	35		2.123	12	36	22	1.831
	13	41	42	1.570			3	29	24		2.086	13	37	25	1.791
	14	42	54	1.531			4	30	14		2.049	14	38	30	1.750
	15	44	11	1.491			5	31	5		2.011	15	39	38	1.708
	16	45	30	1.450			6	31	58		1.973	16	40	48	1.667
	17	46	56	1.406			7	32	52		1.935	17	42	0	1.625
	18	48	26	1.361			8	33	50		1.897	18	43	16	1.581
	19	50	2	1.314			9	34	49		1.860	19	44	36	1.535
	20	51	46	1.263			10	35	48		1.822	20	46	0	1.489
	21	53	42	1.209			11	36	50		1.784	21	47	26	1.441
	22	56	2	1.145			12	37	52		1.745	22	49	2	1.390
23	59	6	1.065	13	38	58	1.705	23	50	38	1.336				
2.10	0	28	26	2.100	14	40	4	1.664	24	52	30	1.280			
	1	29	13	2.064	15	41	14	1.623	25	54	29	1.217			
	2	30	2	2.027	16	42	28	1.582	26	57	2	1.142			
	3	30	52	1.991	17	43	44	1.539	27	60	38	1.046			
	4	31	44	1.955	18	45	4	1.494	2.35	0	25	11	2.350		
	5	32	37	1.918	19	46	28	1.449		1	25	55	2.311		
	6	33	41	1.882	20	47	58	1.403		2	26	42	2.272		
	7	34	28	1.846	21	49	32	1.354		3	27	28	2.234		
	8	35	25	1.809	22	51	14	1.301		4	28	18	2.195		
	9	36	24	1.771	23	53	8	1.246		5	29	8	2.155		
	10	37	26	1.732	24	55	18	1.184		6	29	58	2.116		
	11	38	30	1.695	25	58	2	1.107		7	30	51	2.076		
	12	39	36	1.656	26	62	56	.982		8	31	46	2.037		
	13	40	42	1.617	2.25	0	26	23		2.250	9	32	42	1.997	
	14	41	54	1.577		1	27	6		2.212	10	33	38	1.957	
	15	43	10	1.537		2	27	54		2.173	11	34	38	1.917	
	16	44	28	1.495		3	28	42		2.135	12	35	39	1.876	
17	45	49	1.451							13	36	42	1.834		



TABLE III - Continued

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Continued

M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂		
		(deg)	(min)				(deg)	(min)				(deg)	(min)			
2.35	14	37	46	1.793	2.45	21	44	57	1.560	2.60	26	50	17	1.401		
	15	38	54	1.751		22	46	21	1.510		27	52	6	1.340		
	16	40	2	1.709		23	47	50	1.459		28	54	4	1.272		
	17	41	15	1.666		24	49	25	1.405		29	56	22	1.197		
	18	42	28	1.622		25	51	7	1.349		30	59	22	1.105		
	19	43	45	1.576		26	53	1	1.287		2.70	0	21	44	2.700	
	20	45	6	1.530		27	55	12	1.219			1	22	26	2.652	
	21	46	31	1.483		28	57	52	1.140			2	23	10	2.608	
	22	48	2	1.432		29	61	51	1.023			3	23	56	2.564	
	23	49	38	1.380		2.50	0	23	35			2.500	4	24	43	2.519
	24	51	23	1.324			1	24	19			2.459	5	25	30	2.474
	25	53	16	1.264			2	25	5			2.418	6	26	19	2.430
	26	55	30	1.196			3	25	52			2.376	7	27	10	2.386
	27	58	14	1.116			4	26	38			2.335	8	28	2	2.343
	28	63	0	.980			5	27	26			2.294	9	28	55	2.300
	2.40	0	24	37			2.400	6	28		17	2.252	10	29	50	2.256
		1	25	21			2.360	7	29		8	2.210	11	30	45	2.211
		2	26	7			2.320	8	30		1	2.169	12	31	42	2.166
		3	26	54			2.280	9	30		56	2.126	13	32	42	2.121
		4	27	42			2.241	10	31		52	2.085	14	33	44	2.075
		5	28	32			2.201	11	32		49	2.038	15	34	47	2.029
		6	29	22			2.160	12	33		47	2.000	16	35	52	1.982
		7	30	14			2.120	13	34		48	1.957	17	36	58	1.935
		8	31	8			2.080	14	35		52	1.916	18	38	6	1.887
		9	32	4			2.041	15	36		56	1.871	19	39	17	1.839
		10	33	0			2.000	16	38		5	1.827	20	40	30	1.790
		11	33	58			1.959	17	39		14	1.783	21	41	45	1.740
		12	34	59		1.917	18	40	24		1.739	22	43	3	1.690	
13		36	0	1.876	19	41	37	1.692	23	44	23	1.638				
14		37	6	1.835	20	42	54	1.645	24	45	48	1.584				
15		38	13	1.792	21	44	14	1.598	25	47	19	1.530				
16		39	22	1.748	22	45	35	1.548	26	48	52	1.472				
17		40	32	1.704	23	47	2	1.496	27	50	32	1.412				
18		41	46	1.660	24	48	38	1.443	28	52	20	1.345				
19		43	2	1.615	25	50	18	1.387	29	54	19	1.280				
20		44	20	1.570	26	52	5	1.327	30	56	36	1.202				
21		45	43	1.522	27	54	0	1.263	31	59	38	1.198				
22		47	10	1.472	28	56	20	1.189	2.80	0	20	56	2.800			
23		48	44	1.421	29	59	17	1.100		1	21	38	2.753			
24		50	23	1.367	2.60	0	22	37		2.600	2	22	22	2.707		
25		52	10	1.309		1	23	20		2.556	3	23	6	2.661		
26		54	9	1.243		2	24	6		2.512	4	23	52	2.615		
27		56	32	1.169		3	24	50		2.470	5	24	38	2.569		
28	59	45	1.077	4		25	37	2.426		6	25	26	2.523			
2.45	0	24	5	2.450		5	26	26		2.384	7	26	16	2.476		
	1	24	50	2.408		6	27	15		2.341	8	27	9	2.431		
	2	25	34	2.367		7	28	6		2.299	9	28	2	2.386		
	3	26	21	2.326		8	28	58	2.257	10	28	56	2.339			
	4	27	9	2.286		9	29	52	2.214	11	29	52	2.293			
	5	27	57	2.246	10	30	47	2.173	12	30	50	2.247				
	6	28	48	2.205	11	31	45	2.129	13	31	48	2.200				
	7	29	40	2.163	12	32	43	2.084	14	32	48	2.153				
	8	30	33	2.122	13	33	43	2.039	15	33	50	2.106				
	9	31	27	2.081	14	34	45	1.995	16	34	54	2.057				
	10	32	23	2.041	15	35	48	1.950	17	36	0	2.010				
	11	33	22	2.000	16	36	54	1.906	18	37	8	1.961				
	12	34	22	1.958	17	38	2	1.860	19	38	17	1.912				
	13	35	24	1.917	18	39	11	1.813	20	39	30	1.863				
	14	36	28	1.876	19	40	23	1.767	21	40	44	1.811				
	15	37	34	1.834	20	41	38	1.719	22	42	0	1.757				
	16	38	42	1.790	21	42	55	1.671	23	43	20	1.704				
	17	39	50	1.746	22	44	14	1.621	24	44	42	1.649				
	18	41	2	1.701	23	45	38	1.569	25	46	9	1.594				
	19	42	16	1.655	24	47	6	1.516	26	47	37	1.539				
	20	43	36	1.608	25	48	38	1.459	27	49	12	1.480				



TABLE III - Continued

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Continued

M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂
		(deg)	(min)				(deg)	(min)				(deg)	(min)	
2.80	28	50	54	1.418	3.00	26	45	32	1.659	3.10	22	39	28	1.946
	29	52	42	1.350		27	47	0	1.600		23	40	43	1.890
	30	54	48	1.279		28	48	34	1.539		24	42	0	1.832
	31	57	9	1.198		29	50	14	1.475		25	43	20	1.775
	32	60	25	1.093		30	52	0	1.407		26	44	44	1.715
2.90	0	20	10	2.900	3.05	31	53	57	1.335	3.15	27	46	8	1.654
	1	20	52	2.852		32	56	7	1.254		28	47	40	1.593
	2	21	35	2.804		0	19	8	3.050		29	49	16	1.528
	3	22	19	2.755		1	19	50	3.000		30	50	58	1.462
	4	23	4	2.709		2	20	33	2.949		31	52	46	1.391
	5	23	50	2.660		3	21	18	2.898		32	54	50	1.316
	6	24	38	2.612		4	22	2	2.847		33	57	10	1.230
	7	25	28	2.564		5	22	48	2.796		34	60	12	1.124
	8	26	19	2.517		6	23	36	2.746		0	18	31	3.150
	9	27	13	2.470		7	24	26	2.695		1	19	12	3.096
	10	28	7	2.422		8	25	16	2.643		2	19	54	3.044
	11	29	3	2.375		9	26	8	2.594		3	20	38	2.992
	12	30	0	2.326		10	27	1	2.545		4	21	22	2.940
	13	30	57	2.279		11	27	57	2.495		5	22	7	2.887
	14	31	58	2.231		12	28	53	2.443		6	22	56	2.836
	15	33	0	2.182		13	29	52	2.392		7	23	47	2.784
	16	34	2	2.132		14	30	50	2.341		8	24	37	2.731
	17	35	6	2.083		15	31	50	2.290		9	25	29	2.679
	18	36	13	2.031		16	32	55	2.239		10	26	22	2.627
	19	37	22	1.980		17	34	0	2.186		11	27	18	2.575
	20	38	33	1.929		18	35	6	2.133		12	28	16	2.522
	21	39	46	1.875		19	36	14	2.080		13	29	14	2.469
	22	41	1	1.822		20	37	23	2.027		14	30	12	2.416
	23	42	18	1.768		21	38	35	1.971		15	31	12	2.363
	24	43	40	1.713		22	39	50	1.916		16	32	15	2.310
25	45	4	1.656	23	41	5	1.860	17	33	19	2.255			
26	46	30	1.599	24	42	21	1.803	18	34	24	2.200			
27	48	2	1.541	25	43	42	1.747	19	35	30	2.145			
28	49	39	1.481	26	45	8	1.689	20	36	39	2.088			
29	51	24	1.415	27	46	34	1.629	21	37	50	2.032			
30	53	16	1.346	28	48	7	1.568	22	39	2	1.975			
31	55	22	1.270	29	49	45	1.504	23	40	18	1.918			
32	57	56	1.177	30	51	28	1.436	24	41	35	1.860			
3.00	0	19	28	3.000	31	53	20	1.365	25	42	54	1.802		
	1	20	10	2.949	32	56	10	1.284	26	44	16	1.742		
	2	20	53	2.900	33	58	0	1.195	27	45	41	1.680		
	3	21	37	2.849	34	61	32	1.073	28	47	13	1.618		
	4	22	22	2.800	0	18	49	3.100	29	48	46	1.555		
	5	23	8	2.750	1	19	29	3.046	30	50	27	1.489		
	6	23	56	2.700	2	20	13	2.995	31	52	15	1.419		
	7	24	45	2.651	3	20	58	2.944	32	54	13	1.343		
	8	25	35	2.602	4	21	43	2.892	33	56	28	1.260		
	9	26	27	2.553	5	22	28	2.840	34	59	11	1.160		
	10	27	21	2.503	6	23	16	2.787	35	64	36	.990		
	11	28	17	2.454	7	24	7	2.736	0	18	13	3.200		
	12	29	13	2.404	8	24	58	2.686	1	18	54	3.146		
	13	30	12	2.355	9	25	48	2.635	2	19	36	3.092		
	14	31	12	2.304	10	26	41	2.585	3	20	20	3.040		
	15	32	13	2.254	11	27	37	2.533	4	21	4	2.987		
	16	33	14	2.203	12	28	34	2.483	5	21	50	2.933		
	17	34	19	2.151	13	29	32	2.431	6	22	38	2.879		
	18	35	25	2.100	14	30	31	2.379	7	23	28	2.826		
	19	36	32	2.046	15	31	31	2.327	8	24	19	2.773		
	20	37	43	1.993	16	32	33	2.274	9	25	10	2.720		
	21	38	56	1.939	17	33	36	2.220	10	26	2	2.667		
	22	40	9	1.884	18	34	43	2.166	11	26	58	2.614		
	23	41	25	1.828	19	35	51	2.112	12	27	56	2.561		
	24	42	44	1.773	20	37	1	2.058	13	28	54	2.507		
25	44	7	1.716	21	38	12	2.002	14	29	52	2.452			



TABLE III - Continued

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Continued

M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂		
		(deg)	(min)				(deg)	(min)				(deg)	(min)			
3.20	15	30	52	2.398	3.30	7	22	52	2.910	3.35	36	63	24	1.033		
	16	31	54	2.344		8	23	42	2.856		3.40	0	17	6	3.400	
	17	33	0	2.288		9	24	33	2.801			1	17	49	3.341	
	18	34	5	2.232		10	25	26	2.748			2	18	32	3.284	
	19	35	10	2.176		11	26	23	2.692			3	19	14	3.226	
	20	36	18	2.120		12	27	19	2.636			4	19	58	3.169	
	21	37	32	2.062		13	28	16	2.580			5	20	44	3.111	
	22	38	44	2.005		14	29	16	2.524			6	21	32	3.054	
	23	39	58	1.946		15	30	18	2.467			7	22	21	2.996	
	24	41	16	1.887		16	31	20	2.410			8	23	10	2.937	
	25	42	34	1.829		17	32	22	2.355			9	24	2	2.881	
	26	43	57	1.770		18	33	28	2.297			10	24	55	2.824	
	27	45	22	1.709		19	34	35	2.239			11	25	50	2.766	
	28	46	50	1.646		20	35	43	2.182			12	26	45	2.708	
	29	48	22	1.583		21	36	53	2.123			13	27	44	2.651	
	30	50	0	1.516		22	38	6	2.064			14	28	44	2.595	
	31	51	45	1.445		23	39	21	2.005			15	29	44	2.536	
	32	53	40	1.370		24	40	36	1.945			16	30	46	2.477	
	33	55	50	1.289		25	41	54	1.884			17	31	50	2.420	
	34	58	24	1.196		26	43	13	1.823			18	32	54	2.360	
35	62	6	1.067	27	44	34	1.760	19	34	2		2.301				
3.25	0	17	55	3.250	3.35	29	47	34	1.631	3.45	20	35	10	2.241		
	18	36	3.195	30		49	10	1.565	21		36	19	2.180			
	2	19	18	3.140		31	50	50	1.496		22	37	30	2.120		
	3	20	1	3.086		32	52	40	1.423		23	38	44	2.058		
	4	20	46	3.031		33	54	42	1.344		24	40	0	1.997		
	5	21	30	2.977		34	57	1	1.256		25	41	16	1.935		
	6	22	20	2.923		35	59	54	1.152		26	42	35	1.873		
	7	23	10	2.868		3.40	0	17	22		3.350	27	43	58	1.809	
	8	24	1	2.815			1	18	3		3.294	28	45	24	1.744	
	9	24	52	2.760			2	18	45		3.237	29	46	54	1.677	
	10	25	44	2.706			3	19	28		3.181	30	48	25	1.610	
	11	26	40	2.651			4	20	13		3.125	31	50	4	1.540	
	12	27	36	2.597			5	21	0		3.068	32	51	50	1.468	
	13	28	34	2.543			6	21	49		3.013	33	53	42	1.392	
	14	29	34	2.489			7	22	37		2.955	34	55	53	1.310	
	15	30	35	2.433			8	23	26		2.899	35	58	24	1.214	
	16	31	38	2.378			9	24	18		2.842	36	61	54	1.090	
	17	32	41	2.322			3.45	10	25		10	2.786	0	16	51	3.450
	18	33	46	2.265				11	26		6	2.729	1	17	32	3.391
	19	34	53	2.210				12	27		4	2.672	2	18	13	3.333
20	36	1	2.152	13	28			2	2.615	3	18	56	3.275			
21	37	12	2.094	14	29			0	2.558	4	19	41	3.216			
22	38	24	2.035	15	30			0	2.500	5	20	26	3.156			
23	39	38	1.975	16	31			3	2.442	6	21	14	3.097			
24	40	54	1.916	17	32			6	2.385	7	22	4	3.040			
25	42	10	1.856	18	33			12	2.327	8	22	54	2.981			
26	43	31	1.795	19	34			19	2.269	9	23	46	2.922			
27	44	55	1.734	20	35	25		2.211	10	24	39	2.864				
28	46	25	1.671	21	36	34		2.152	11	25	34	2.805				
29	47	57	1.607	22	37	46		2.092	12	26	30	2.747				
30	49	33	1.540	23	39	2		2.032	13	27	27	2.687				
31	51	18	1.469	24	40	18		1.970	14	28	26	2.630				
32	53	8	1.398	25	41	35		1.910	15	29	28	2.570				
33	55	14	1.318	26	42	54		1.846	16	30	30	2.512				
34	57	38	1.228	27	44	18		1.783	17	31	34	2.452				
35	60	50	1.116	28	45	44		1.719	18	32	37	2.391				
3.30	0	17	35	3.300	29	47		14	1.655	19	33	45	2.331			
	1	18	18	3.244	30	48	46	1.587	20	34	51	2.270				
	2	19	2	3.189	31	50	28	1.520	21	36	0	2.208				
	3	19	46	3.132	32	52	14	1.445	22	37	14	2.147				
	4	20	30	3.077	33	54	10	1.369	23	38	27	2.085				
	5	21	15	3.022	34	56	24	1.283	24	39	43	2.022				
	6	22	4	2.966	35	59	4	1.186	25	41	0	1.958				



TABLE III - Continued

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Continued

M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂
		(deg)	(min)				(deg)	(min)				(deg)	(min)	
3.45	26	42	18	1.895	3.55	16	29	59	2.576	3.65	4	18	41	3.396
	27	43	38	1.831		17	31	3	2.516		5	19	28	3.333
	28	45	5	1.766		18	32	6	2.455		6	20	15	3.270
	29	46	34	1.700		19	33	12	2.392		7	21	4	3.207
	30	48	6	1.631		20	34	20	2.330		8	21	55	3.145
	31	49	42	1.562		21	35	30	2.266		9	22	46	3.082
	32	51	26	1.491		22	36	42	2.203		10	23	40	3.020
	33	53	17	1.416		23	37	55	2.139		11	24	34	2.957
	34	55	22	1.334		24	39	10	2.075		12	25	30	2.894
	35	57	45	1.242		25	40	26	2.009		13	26	28	2.831
	36	61	0	1.127		26	41	44	1.943		14	27	28	2.766
	3.50	0	16	36		3.500	27	43	8		1.877	15	28	28
1		17	16	3.440	28	44	32	1.812	16	29	30	2.640		
2		17	58	3.381	29	45	57	1.745	17	30	34	2.577		
3		18	40	3.321	30	47	28	1.675	18	31	38	2.513		
4		19	24	3.262	31	49	2	1.606	19	32	46	2.448		
5		20	10	3.203	32	50	44	1.535	20	33	53	2.383		
6		21	0	3.141	33	52	30	1.459	21	35	2	2.317		
7		21	48	3.082	34	54	29	1.380	22	36	14	2.252		
8		22	38	3.023	35	56	42	1.291	23	37	26	2.187		
9		23	29	2.963	36	59	26	1.187	24	38	40	2.121		
10		24	24	2.904	37	64	10	1.019	25	39	56	2.055		
11		25	18	2.844	3.60	0	16	8	3.600	26	41	14	1.987	
12	26	14	2.785	1		16	47	3.538	27	42	34	1.922		
13	27	10	2.724	2		17	29	3.474	28	43	57	1.855		
14	28	10	2.662	3		18	12	3.412	29	45	24	1.786		
15	29	11	2.602	4		18	56	3.350	30	46	54	1.718		
16	30	14	2.543	5		19	41	3.288	31	48	26	1.648		
17	31	17	2.483	6		20	30	3.226	32	50	4	1.575		
18	32	22	2.422	7		21	19	3.166	33	51	49	1.500		
19	33	28	2.361	8		22	10	3.104	34	53	44	1.422		
20	34	35	2.299	9		23	0	3.044	35	55	46	1.337		
21	35	45	2.237	10		23	54	2.981	36	58	16	1.241		
22	36	57	2.174	11		24	48	2.919	37	60	50	1.117		
23	38	10	2.111	12	25	44	2.857	3.70	0	15	41	3.700		
24	39	24	2.048	13	26	42	2.795		1	16	20	3.635		
25	40	42	1.983	14	27	41	2.731		2	17	2	3.569		
26	42	2	1.918	15	28	41	2.670		3	17	44	3.503		
27	43	24	1.852	16	29	45	2.608		4	18	28	3.438		
28	44	47	1.787	17	30	47	2.546		5	19	15	3.376		
29	46	14	1.720	18	31	52	2.483		6	20	2	3.312		
30	47	46	1.652	19	32	58	2.419		7	20	50	3.250		
31	49	21	1.584	20	34	6	2.357		8	21	41	3.186		
32	51	4	1.513	21	35	15	2.292		9	22	33	3.124		
33	52	53	1.437	22	36	27	2.227		10	23	26	3.060		
34	54	55	1.356	23	37	39	2.163		11	24	21	2.996		
35	57	14	1.266	24	38	53	2.097	12	25	17	2.932			
36	60	12	1.158	25	40	10	2.032	13	26	14	2.868			
3.55	0	16	22	3.550	26	41	28	1.966	14	27	14	2.802		
	1	17	2	3.488	27	42	49	1.900	15	28	15	2.737		
	2	17	44	3.426	28	44	13	1.834	16	29	16	2.672		
	3	18	26	3.365	29	45	39	1.766	17	30	20	2.606		
	4	19	10	3.304	30	47	8	1.699	18	31	26	2.540		
	5	19	56	3.244	31	48	43	1.628	19	32	32	2.475		
	6	20	46	3.184	32	50	24	1.556	20	33	40	2.410		
	7	21	34	3.125	33	52	9	1.481	21	34	49	2.342		
	8	22	25	3.065	34	54	7	1.402	22	36	0	2.277		
	9	23	15	3.005	35	56	14	1.315	23	37	13	2.211		
	10	24	10	2.943	36	58	49	1.215	24	38	28	2.145		
	11	25	4	2.882	37	62	39	1.075	25	39	42	2.077		
12	26	0	2.821	3.65	0	15	54	3.650	26	41	2	2.010		
13	26	56	2.760		1	16	34	3.587	27	42	22	1.943		
14	27	54	2.699		2	17	16	3.523	28	43	45	1.875		
15	28	56	2.637		3	17	58	3.460	29	45	10	1.806		
									30	46	38	1.737		



TABLE III - Concluded

VALUES OF MACH NUMBER, DEVIATION OF FLOW, ANGLE OF SHOCK,
AND MACH NUMBER BEHIND THE SHOCK - Concluded

M ₁	δ (deg)	ε		M ₂	M ₁	δ (deg)	ε		M ₂
		(deg)	(min)				(deg)	(min)	
3.70	31	48	10	1.665	3.90	15	27	25	2.864
	32	49	46	1.594		16	28	26	2.796
	33	51	30	1.518		17	29	30	2.727
	34	53	21	1.440		18	30	34	2.657
	35	55	22	1.357		19	31	40	2.587
	36	57	46	1.263		20	32	48	2.517
	37	60	50	1.148		21	33	57	2.446
3.80	0	15	15	3.800	4.0	22	35	8	2.377
	1	15	55	3.734		23	36	21	2.308
	2	16	36	3.665		24	37	36	2.237
	3	17	18	3.597		25	38	50	2.166
	4	18	2	3.532		26	40	8	2.095
	5	18	49	3.467		27	41	28	2.025
	6	19	36	3.402		28	42	50	1.955
	7	20	25	3.336		29	44	13	1.885
	8	21	16	3.270		30	45	40	1.812
	9	22	8	3.204		31	47	10	1.741
	10	23	1	3.137		32	48	43	1.666
	11	23	57	3.069		33	50	22	1.591
	12	24	53	3.002		34	52	6	1.513
	13	25	50	2.935		35	54	2	1.431
	14	26	19	2.866		36	56	9	1.343
	15	27	50	2.800		37	58	38	1.240
	16	28	50	2.733		38	62	8	1.114
	17	29	56	2.665		0	14	29	4.000
	18	31	2	2.598		1	15	10	3.927
	19	32	7	2.531		2	15	50	3.854
	20	33	15	2.462		3	16	33	3.783
21	34	23	2.395	4	17	17	3.711		
22	35	34	2.328	5	18	2	3.639		
23	36	46	2.260	6	18	50	3.568		
24	38	0	2.193	7	19	39	3.497		
25	39	16	2.124	8	20	30	3.426		
26	40	34	2.055	9	21	20	3.355		
27	41	54	1.986	10	22	13	3.285		
28	43	16	1.916	11	23	9	3.213		
29	44	41	1.846	12	24	6	3.141		
30	46	8	1.776	13	25	3	3.070		
31	47	38	1.706	14	26	2	2.999		
32	49	13	1.631	15	27	4	2.926		
33	50	54	1.557	16	28	6	2.856		
34	52	41	1.480	17	29	9	2.780		
35	54	42	1.397	18	30	14	2.713		
36	56	57	1.306	19	31	18	2.641		
37	59	40	1.200	20	32	26	2.570		
38	64	14	1.026	21	33	37	2.497		
3.90	0	14	51	3.900	22	34	46	2.424	
	1	15	30	3.828	23	35	58	2.354	
	2	16	12	3.757	24	37	12	2.282	
	3	16	54	3.687	25	38	28	2.210	
	4	17	37	3.618	26	39	46	2.137	
	5	18	22	3.551	27	41	5	2.065	
	6	19	10	3.482	28	42	26	1.992	
	7	20	0	3.413	29	43	48	1.922	
	8	20	51	3.345	30	45	14	1.850	
	9	21	43	3.276	31	46	42	1.777	
	10	22	36	3.207	32	48	14	1.700	
	11	23	32	3.139	33	49	55	1.625	
	12	24	29	3.070	34	51	38	1.548	
	13	25	26	3.002	35	53	27	1.465	
14	26	24	2.932	36	55	30	1.380		
				37	57	52	1.281		
				38	60	50	1.165		



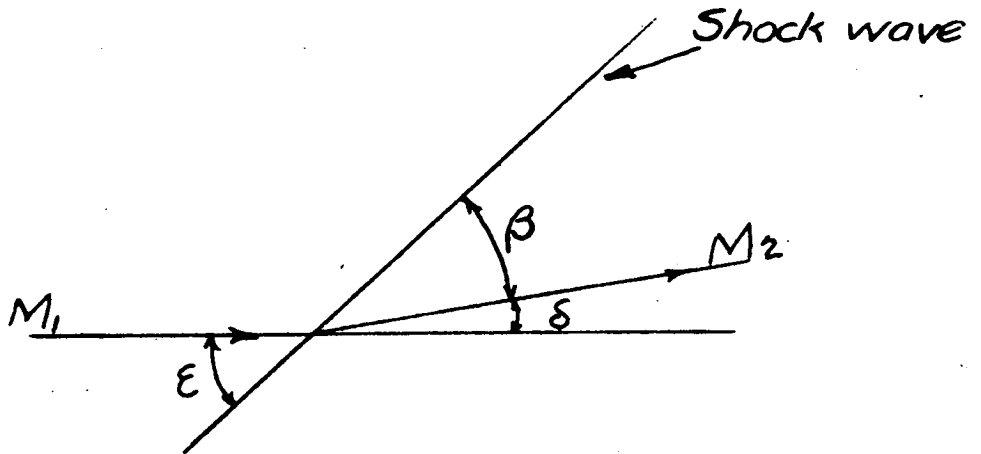
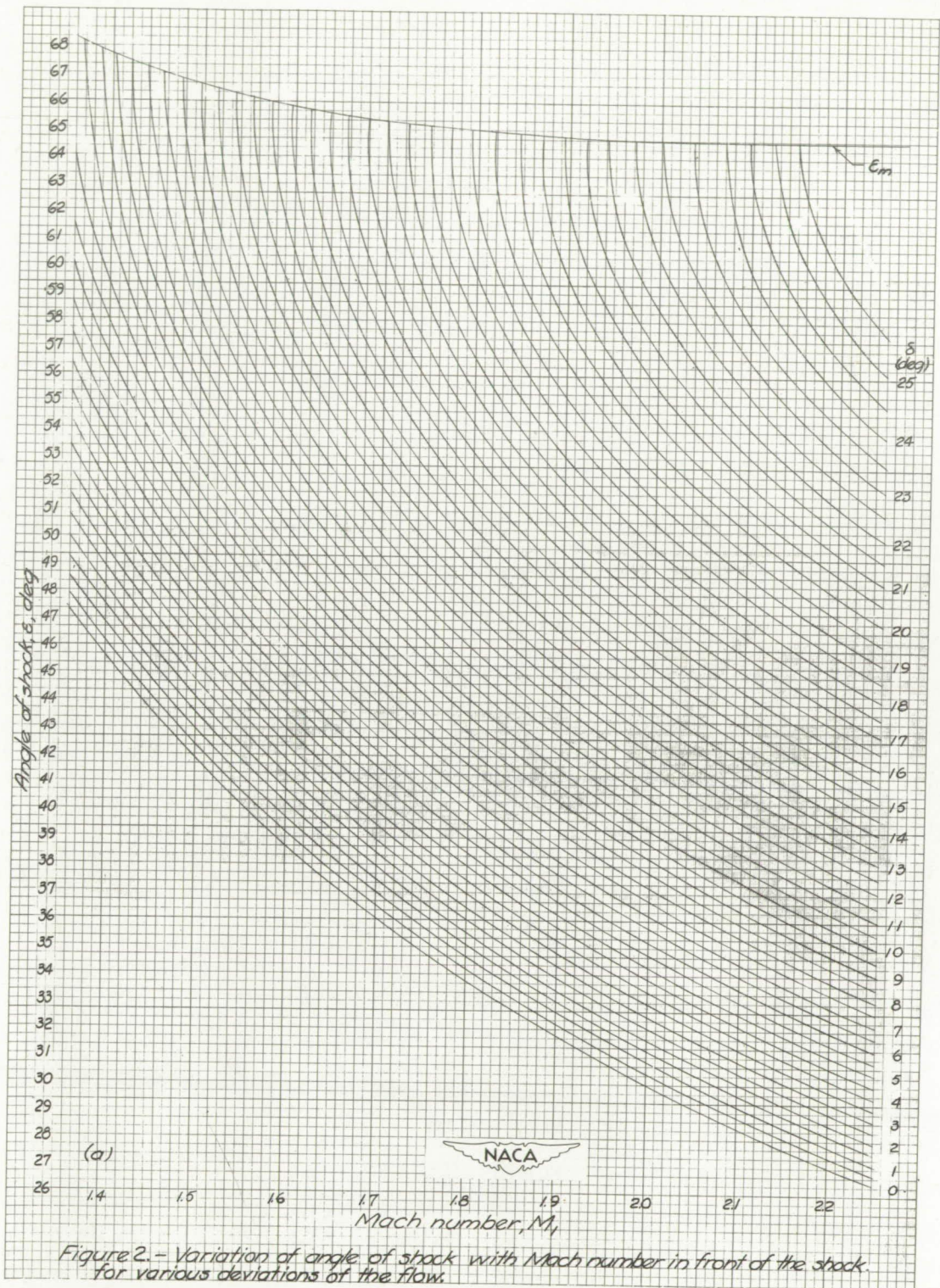
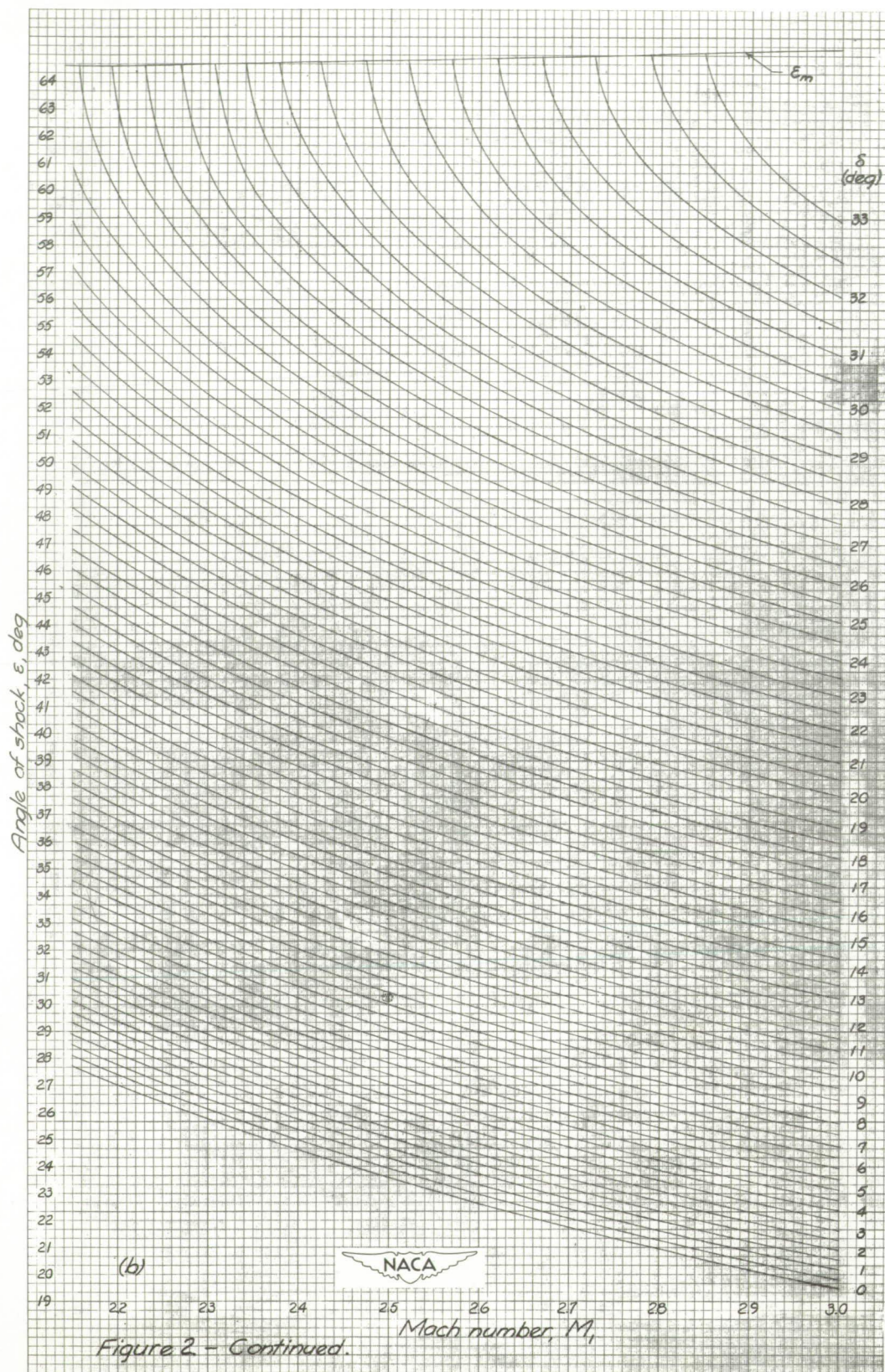


Figure 1.- Diagram showing the angle of the shock, deviation of the flow, and the angle of the flow behind the shock.





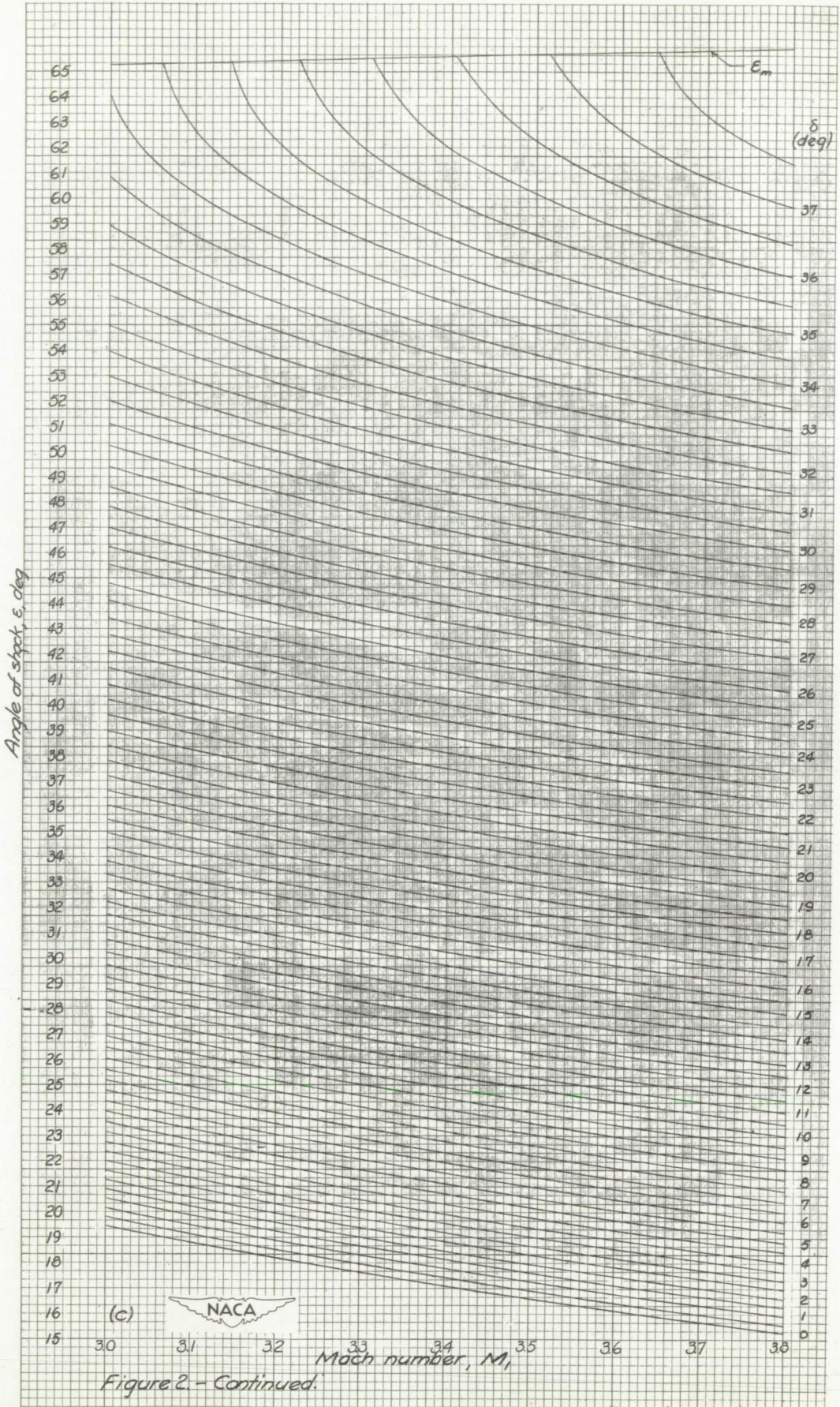


Figure 2. - Continued.

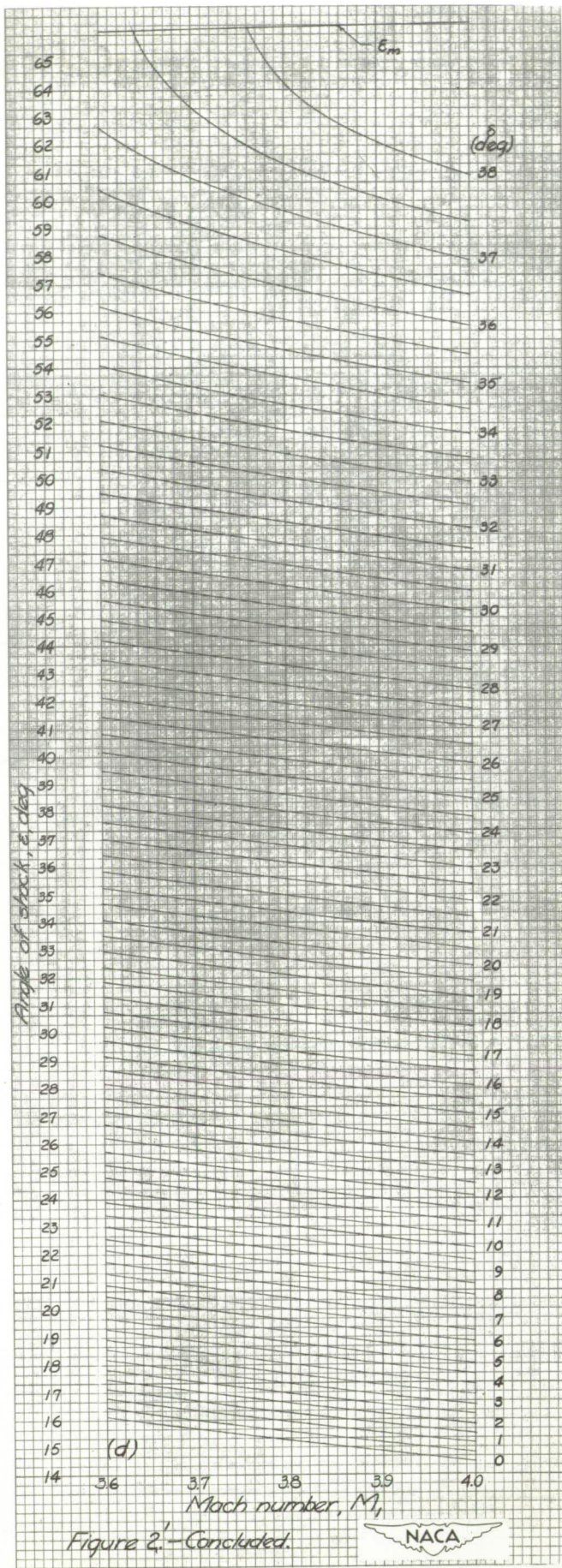


Figure 2'-Concluded.



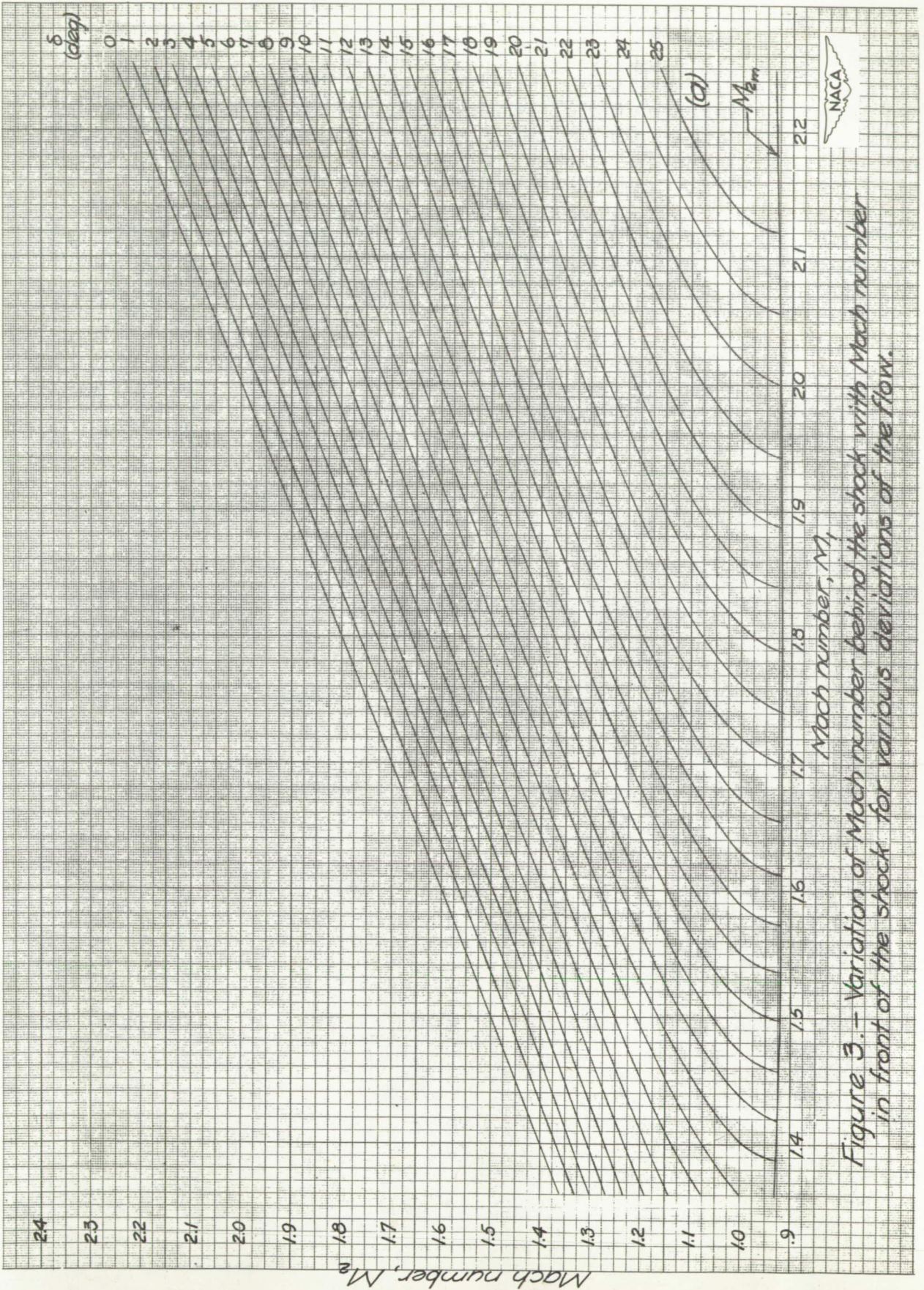


Figure 3. - Variation of Mach number behind the shock with Mach number in front of the shock for various deviations of the flow.

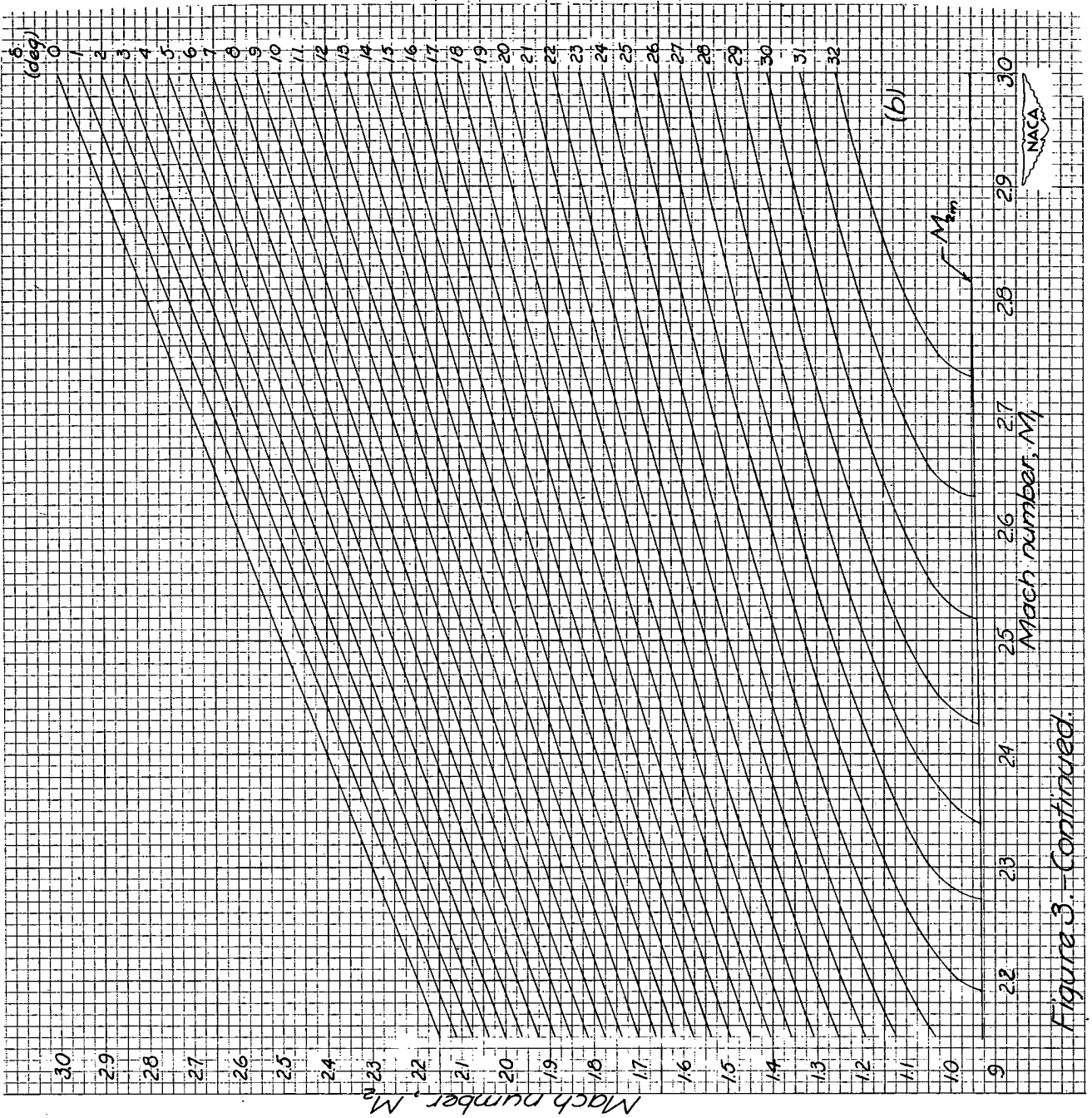


Figure 3. Continued.

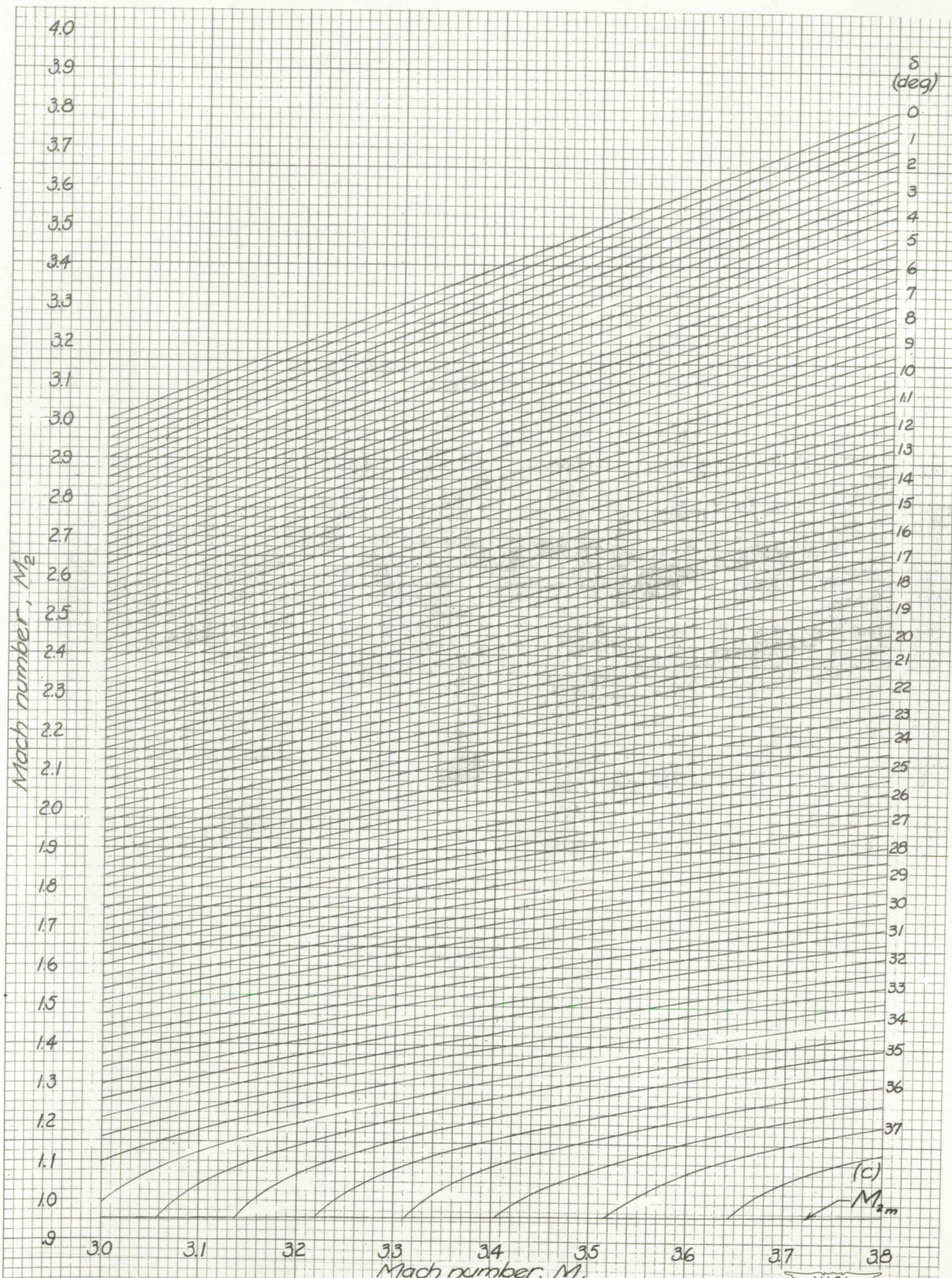
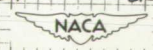


Figure 3. - Continued.



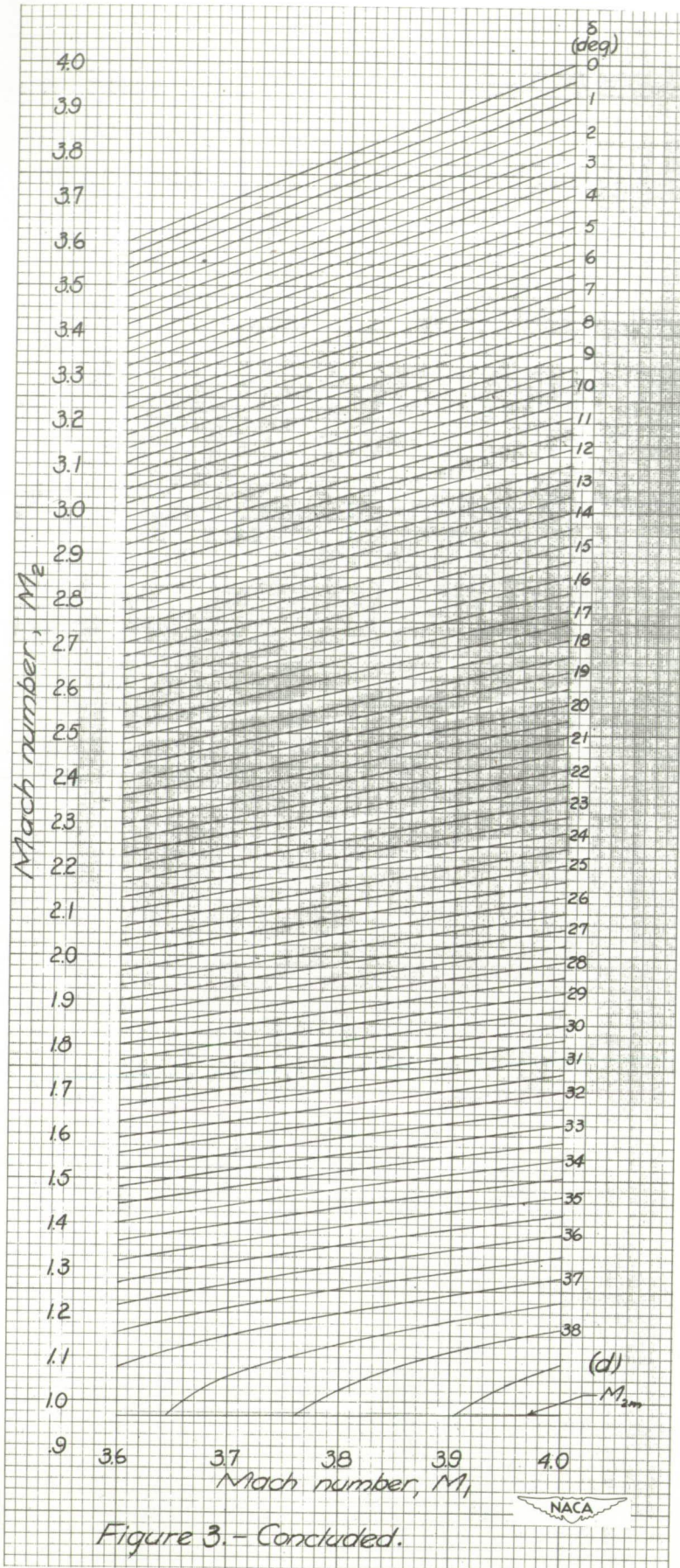


Figure 3.- Concluded.



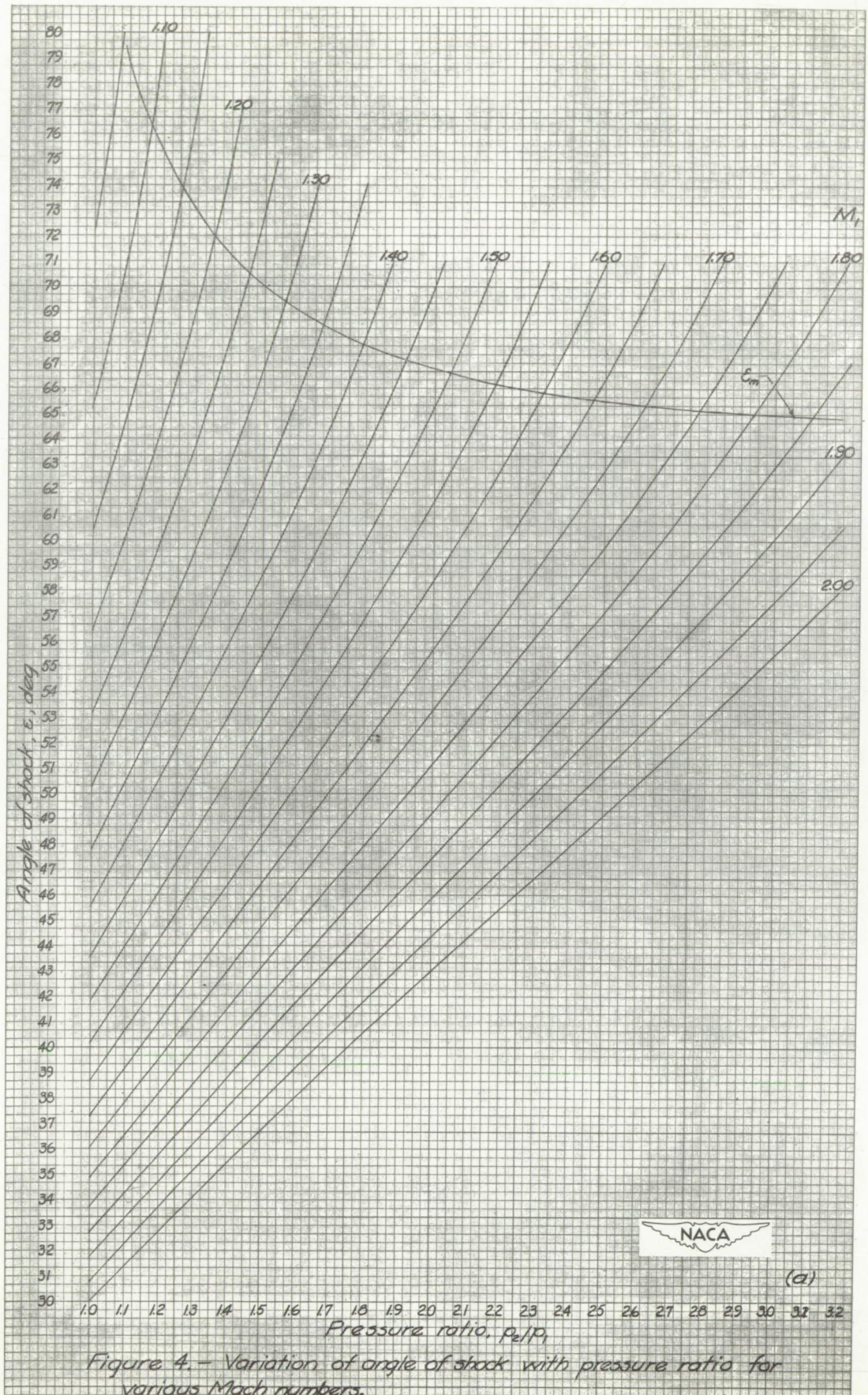


Figure 4. - Variation of angle of shock with pressure ratio for various Mach numbers.

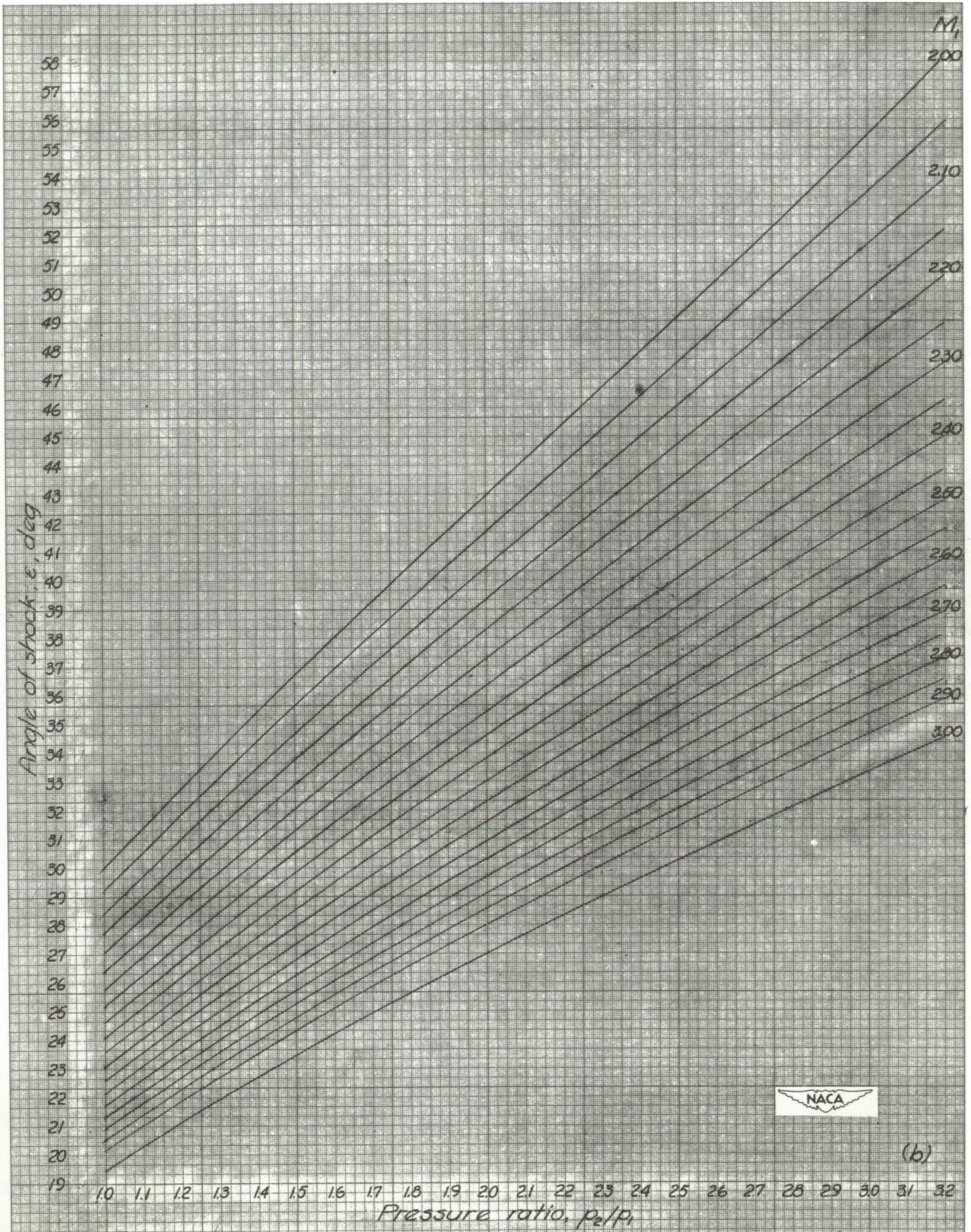


Figure 4.-Continued.



(b)

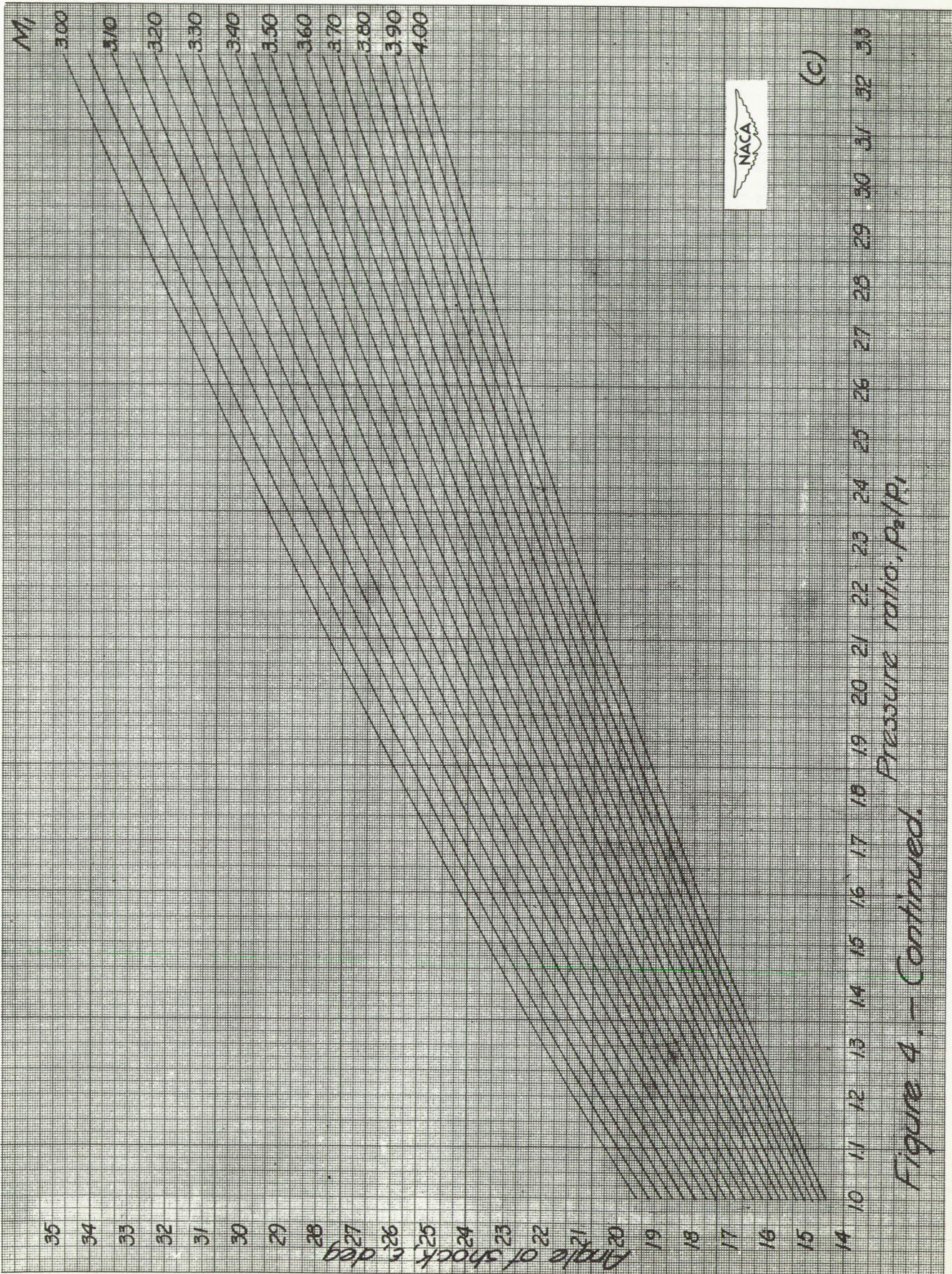


Figure 4. - Continued.

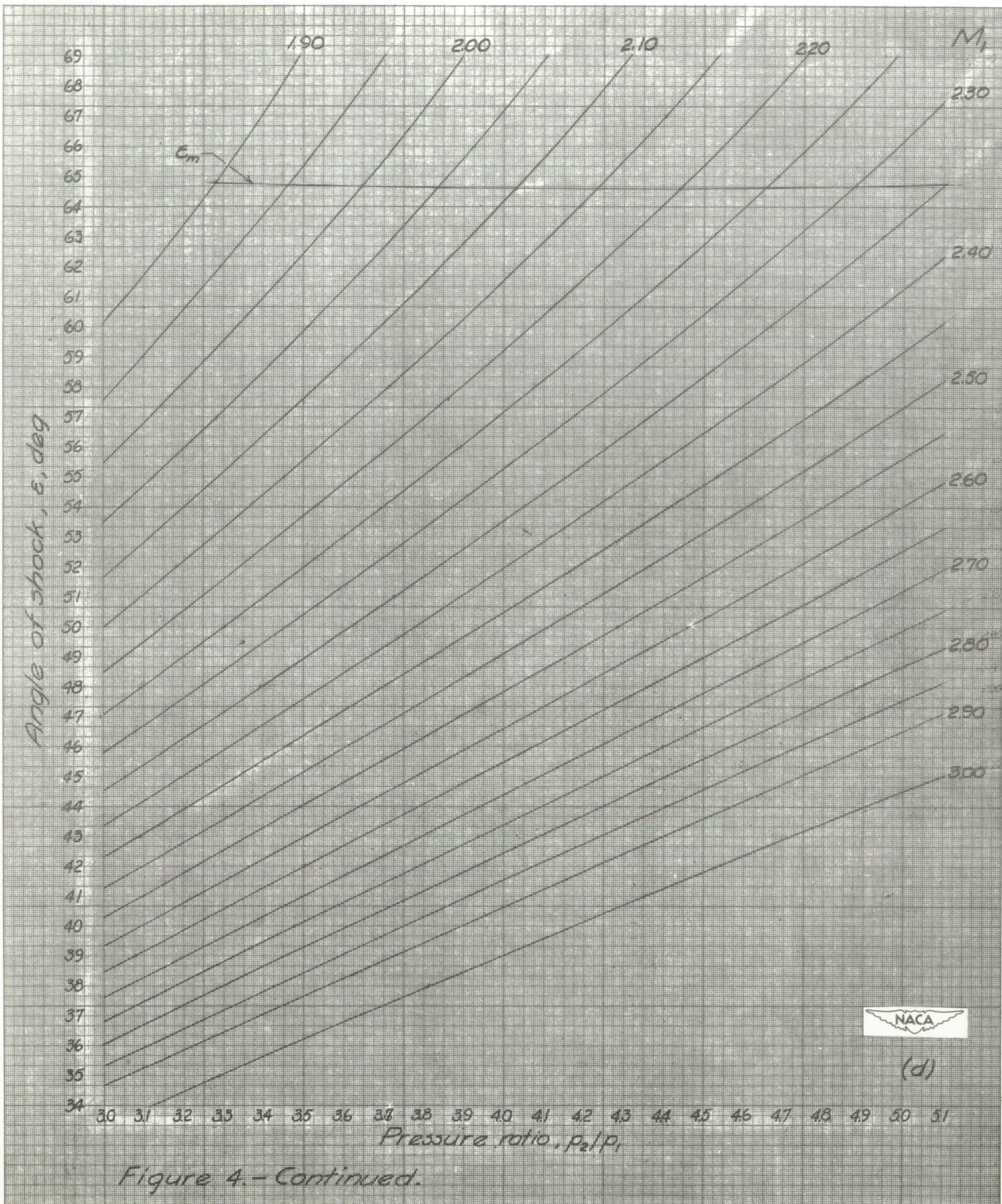


Figure 4. - Continued.

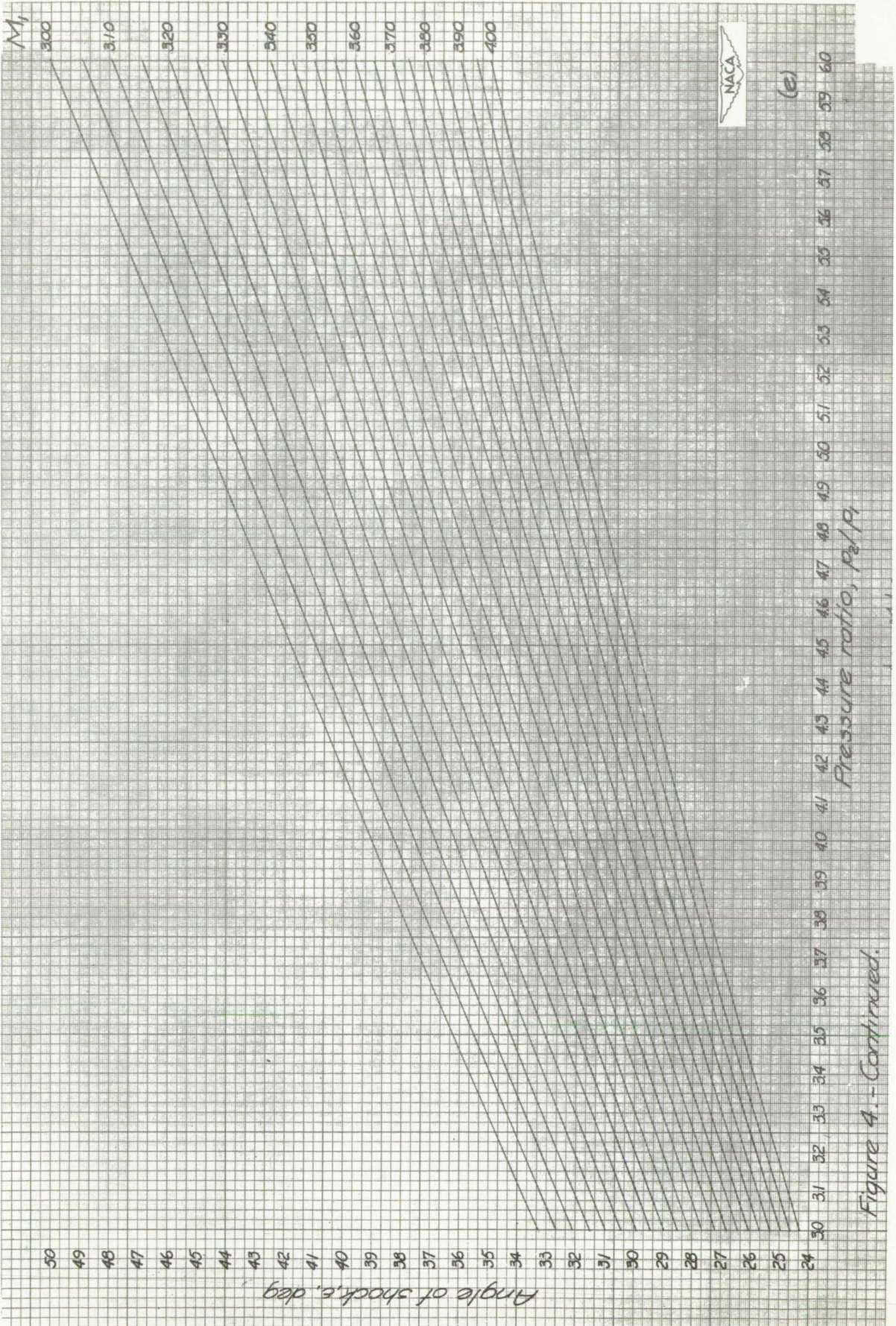


Figure 4. - Continued.

Pressure ratio, P_2/P_1

(e)

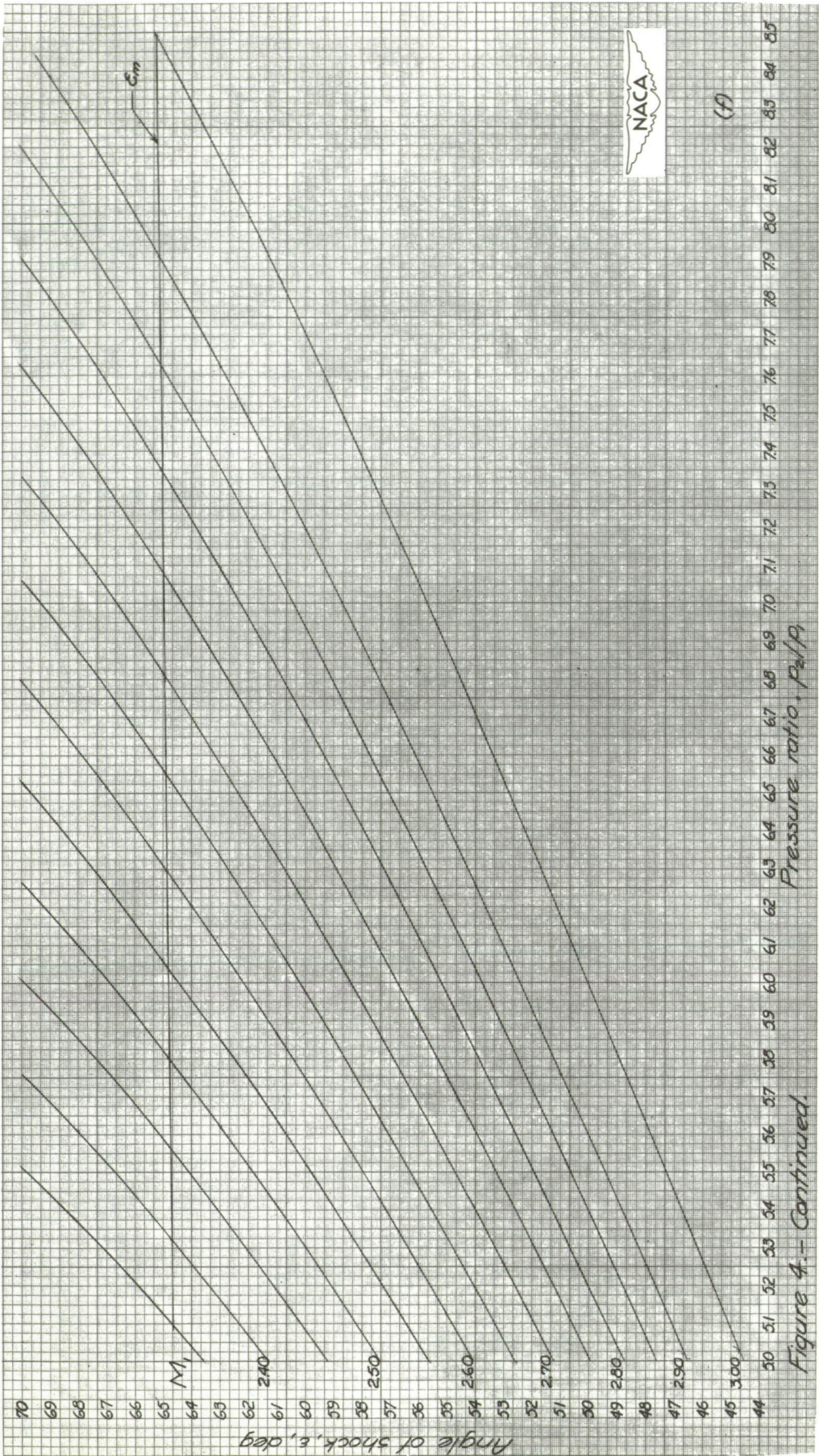


Figure 4.- Continued

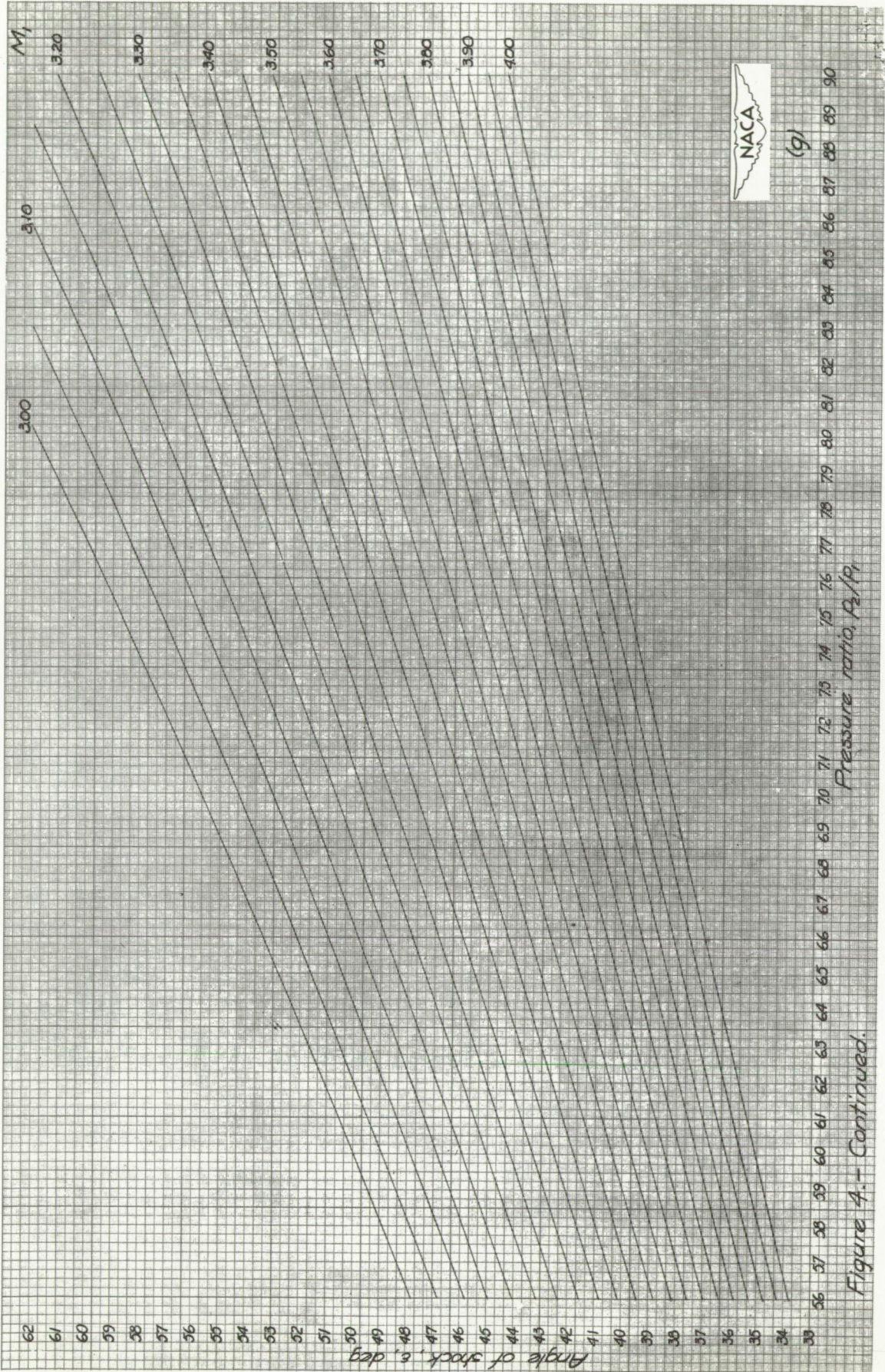


Figure 4.- Continued.

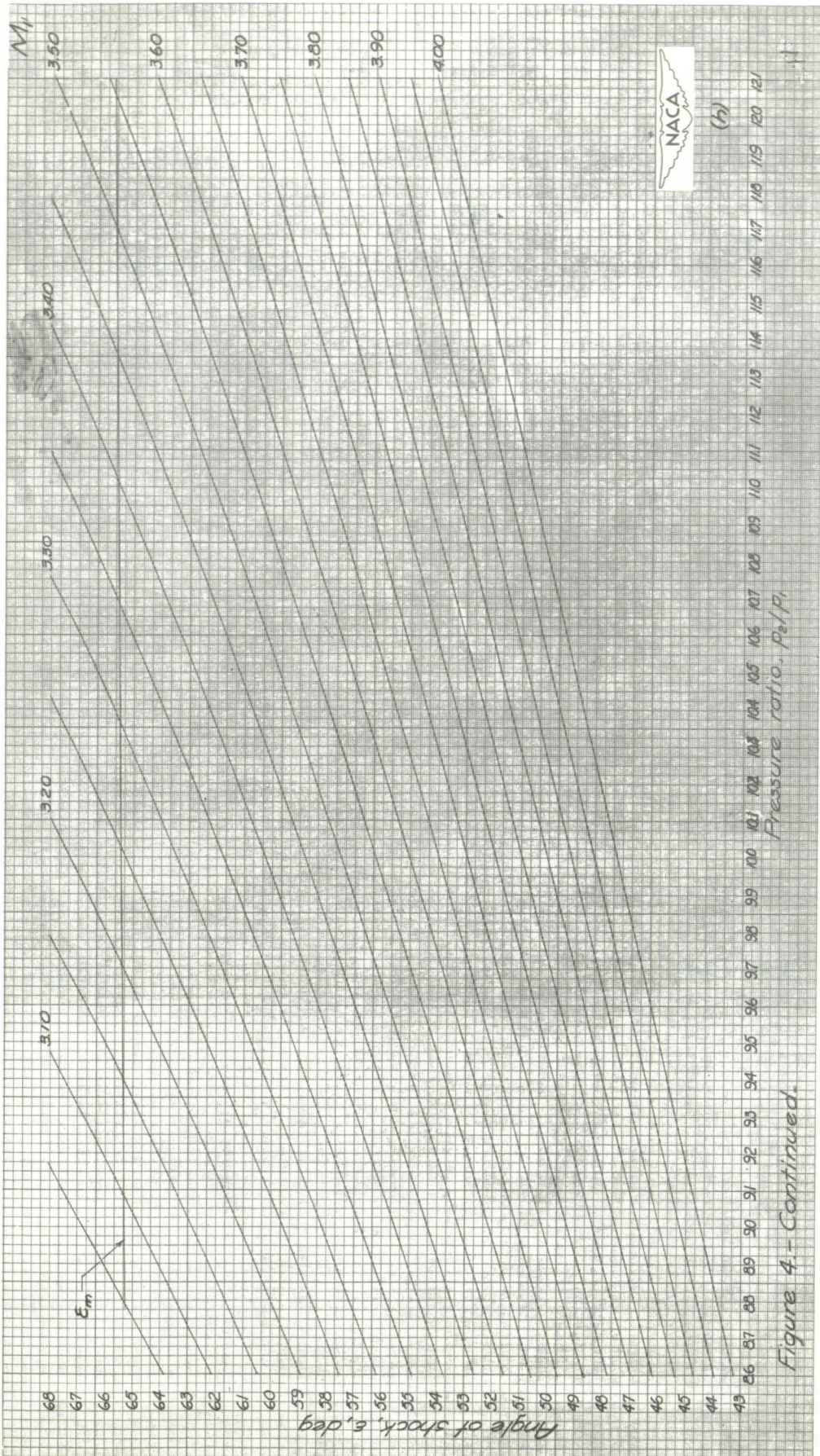


Figure 4.-Continued.

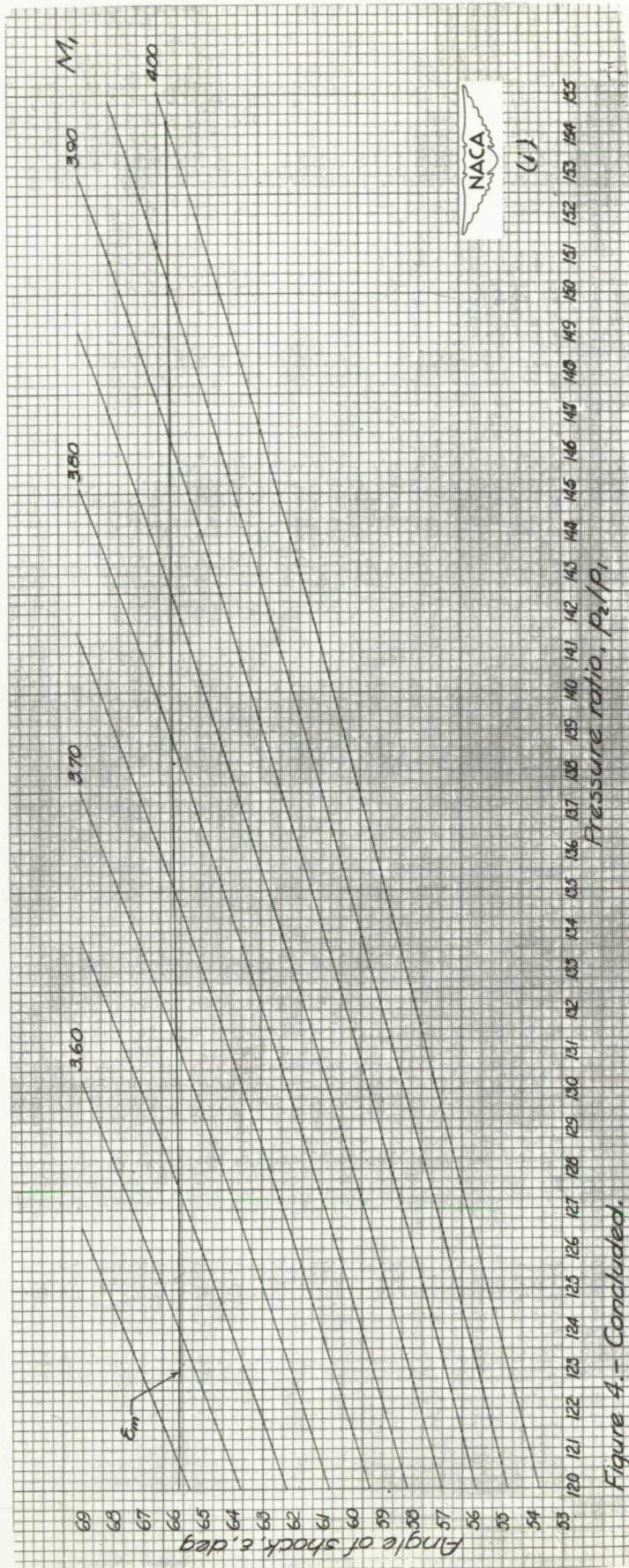


Figure 4.- Concluded.

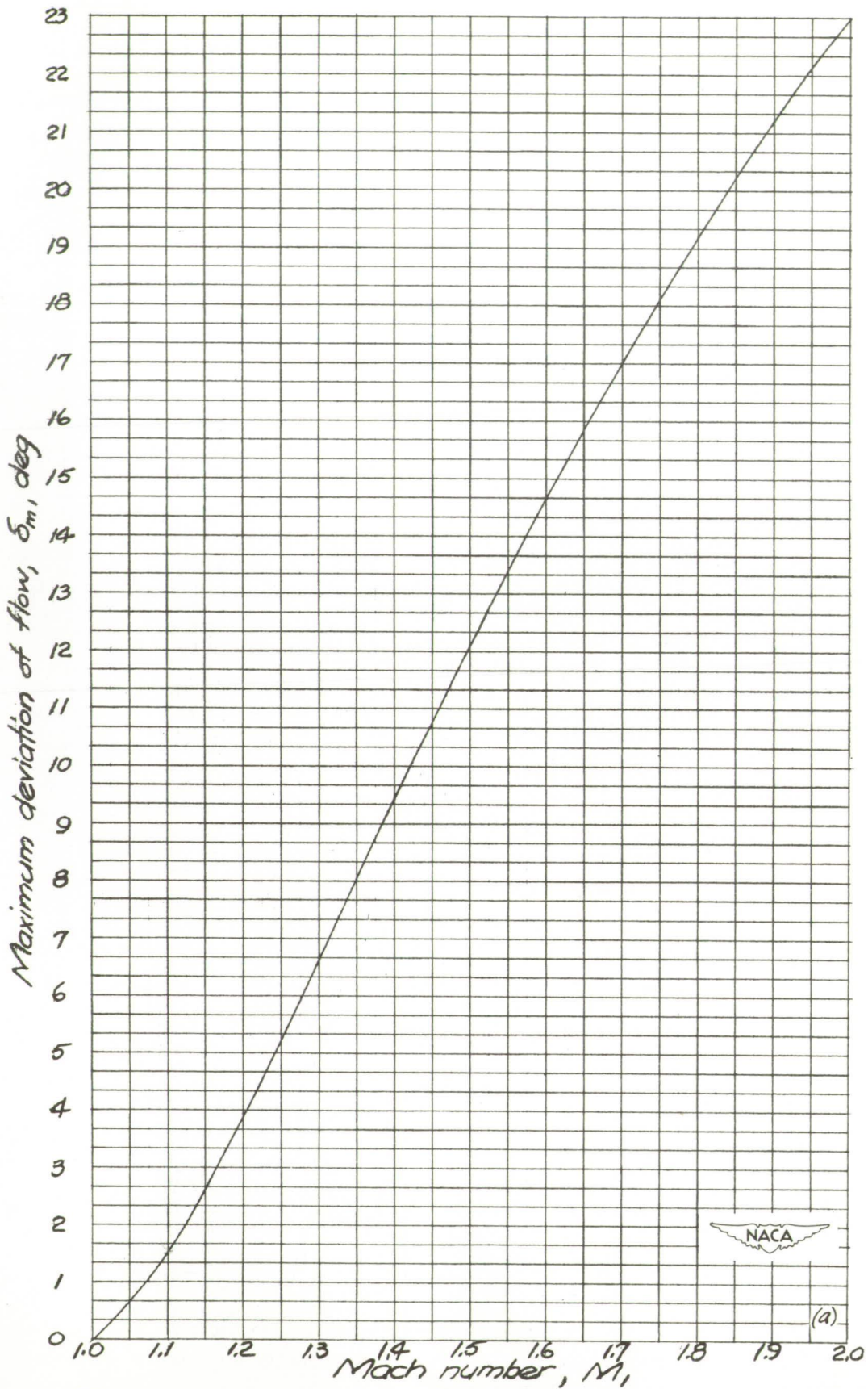


Figure 5.- Variation of maximum deviation of flow across the shock with Mach number in front of shock.

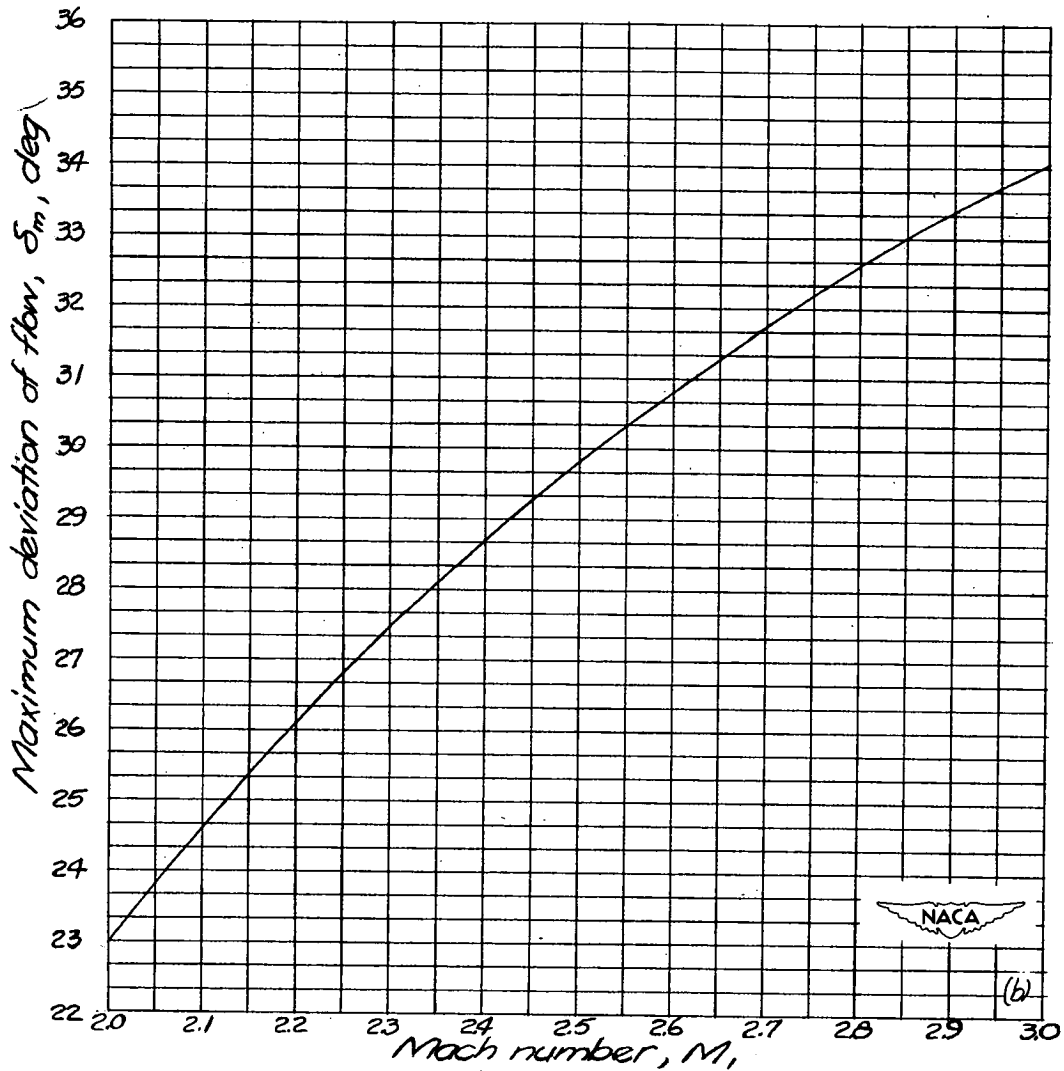


Figure 5.-Continued.

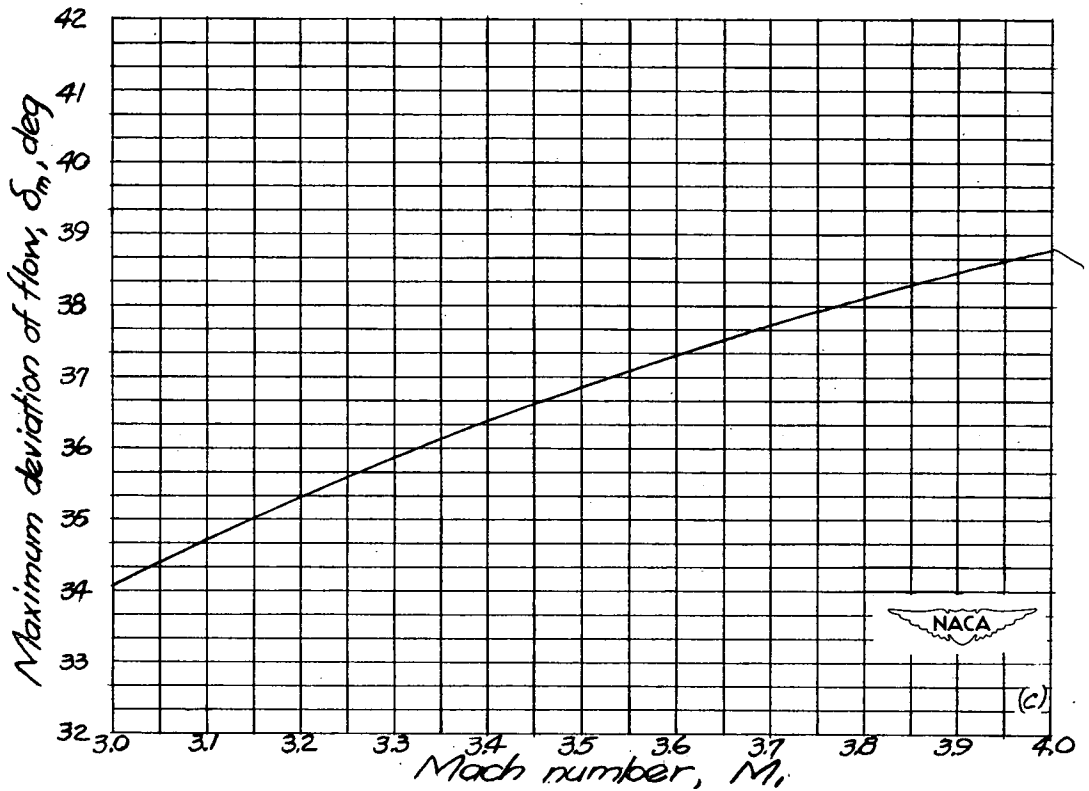
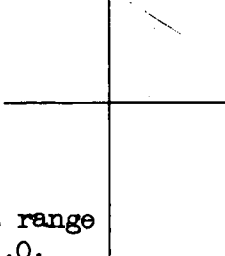


Figure 5.- Concluded.



Abstract

Shock-wave equations have been evaluated for a range of Mach number in front of the shock from 1.05 to 4.0. Mach number behind the shock, pressure ratio, deviation of flow, and angle of shock are presented on charts. Values are also included for density ratio and change in entropy.

Flow, Supersonic

1.1.2.3
S



Tables and Charts of Flow Parameters across Oblique Shocks.

By Mary M. Neice

NACA TN No. 1673

August 1948

(Abstract on Reverse Side)

