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# RESEARCH MEMORANDUM

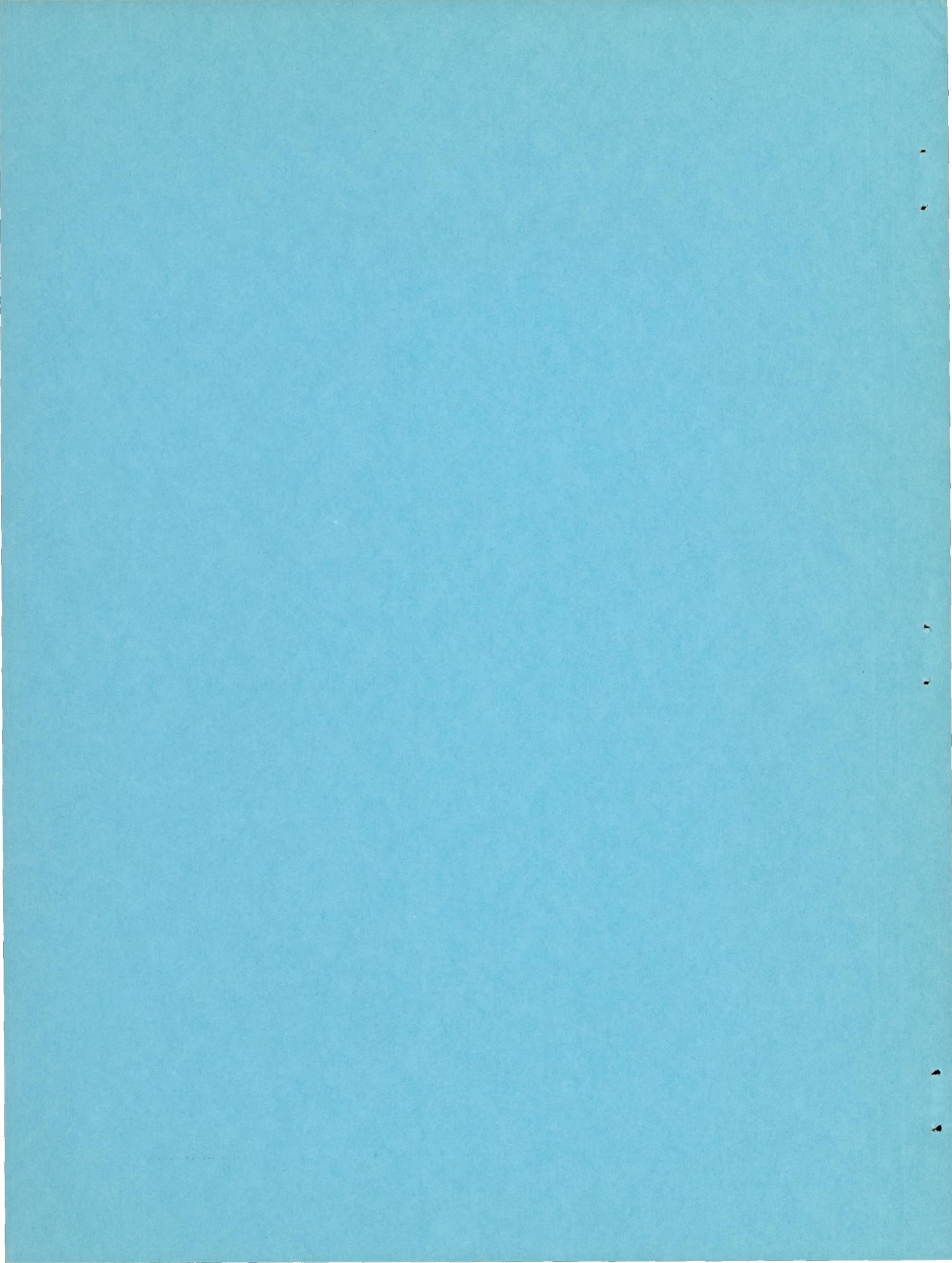
THE VARIATION WITH WING ASPECT RATIO  
OF FLAP EFFECTIVENESS ON THIN RECTANGULAR  
WINGS AT TRANSONIC SPEEDS

By John G. Lowry and Robert T. Taylor

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NATIONAL ADVISORY COMMITTEE  
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SUMMARY

A wind-tunnel investigation has been made in the Langley high-speed 7- by 10-foot tunnel by use of the transonic-bump technique to study the effectiveness of full-span flap-type controls on 31 unswept rectangular wings. Plain flaps with flap-chord ratios of 0.1, 0.2, 0.3, and 0.4 were tested on wings of aspect ratios from 1 to 6 at Mach numbers from 0.4 to 1.1. The data for the most part are presented without analysis.

INTRODUCTION

The variation of control effectiveness with wing aspect ratio and flap-chord ratio is well known at low subsonic speeds (refs. 1 to 5) and at supersonic speeds (refs. 6 and 7). There is, however, very little known about these effects at transonic speeds. In addition there are only a few systematic data available for controls on the thinner (4- to 6-percent thick) airfoils at transonic speeds.

The present paper presents the results of a wind-tunnel investigation to determine flap-effectiveness parameters using 31 small-scale rectangular semispan wings equipped with full-span plain flaps. The transonic speeds were obtained by using the transonic-bump technique in the Langley high-speed 7- by 10-foot tunnel. The variables investigated were wing thickness (4 and 6 percent), wing aspect ratio (aspect ratios from 1 to 6), and flap-chord ratio (flap-chord to wing-chord ratios of 0.1 to 0.4).

In order to expedite the publication of the results, no detailed analysis or discussion of the data will be made. All of the data are presented in tabulated form and in addition some data showing significant trends are presented in graphic form.

## SYMBOLS

$C_L$	lift coefficient, $\frac{\text{Twice semispan lift}}{qS}$
$C_D$	drag coefficient, $\frac{\text{Twice semispan drag}}{qS}$
$C_m$	pitching-moment coefficient about 0.25c, $\frac{\text{Twice semispan pitching moment}}{qSc}$
$C_l$	rolling-moment coefficient, $\frac{\text{Semispan rolling moment}}{qSb}$
$C_n$	yawing-moment coefficient, $\frac{\text{Semispan yawing moment}}{qSb}$
b	wing span, ft
c	wing chord, ft
$c_f$	flap chord, ft
S	wing area, sq ft
t	wing thickness, ft
A	wing aspect ratio, $\frac{b^2}{S}$
q	free-stream dynamic pressure, $\frac{1}{2}\rho V^2$ , lb/sq ft
V	free-stream velocity, ft/sec
$\rho$	free-stream density, slugs/cu ft
R	Reynolds number based on wing chord
M	free-stream Mach number
$M_l$	local Mach number
$\alpha$	angle of attack, deg
$\delta$	flap deflection, deg

$\alpha_0$  flap-effectiveness parameter, effective change in wing angle of attack caused by unit angular change in control-surface deflection

$C_{L\alpha}$  lift-curve slope,  $\frac{\partial C_L}{\partial \alpha}$

## MODELS

The geometric characteristics of the models used in the investigation are given in figure 1. The models were machined from solid steel to either NACA 65A004 or NACA 65A006 airfoil sections. The basic models had no twist or camber and had a taper ratio of 1. The aspect ratio was varied by cutting the wings at the appropriate spanwise station and filing the tip normal to the chord plane.

The flaps were machined integrally with the wing at a deflection of approximately  $10^\circ$ . The flap chords and actual deflections are given on figure 1.

## TESTS

The tests were made by using the transonic-bump technique in the Langley high-speed 7- by 10-foot tunnel. The models were attached to a five-component electrical-strain-gage balance beneath the bump surface. The tests were made over a Mach number range from 0.4 to 1.1 at Reynolds numbers varying from  $0.5 \times 10^6$  to  $1.5 \times 10^6$  (fig. 2). The variation of the local Mach number over the bump in the vicinity of the model is shown in figure 3.

The test angles of attack varied from  $-10^\circ$  to  $25^\circ$  whenever the loads encountered did not exceed the design limit of the balance. The aspect ratio varied from 6 to 2 on the 6-percent-thick wings and from 4 to 1 on the 4-percent-thick wings. Flap chords varied from 0 to  $0.4c$ .

The data have not been corrected for jet-boundary effects on blocking since the models were sufficiently small with respect to tunnel boundaries to make the corrections negligible. No corrections were applied to account for flap deflection under load since checks indicated these too were small. The roll and yaw data presented represent the rolling- and yawing-moment coefficients resulting from deflection of the control on one wing. Since no reflection-plane corrections have been applied to the data, they represent symmetrically deflected controls and

should be reduced if applied to antisymmetric deflection. The magnitude of the corrections (reflection plane) at  $M = 0$  obtained from references 3 and 4 are given in figure 4. The variation of the correction with Mach number has not been established in the transonic-speed range but does decrease to 0 at supersonic speeds.

Because of the small physical size of the models and the resulting inaccuracies in the measurement of the forces and moments, care should be taken in analyzing the tabulated data especially at the lower Mach numbers.

#### RESULTS AND DISCUSSION

The force and moment data obtained in this investigation are presented in tables 1 to 6.

A comparison of the lift-curve slope  $C_{L\alpha}$  with theory is given in figure 5 at  $M = 0.4$ . The variation of  $C_{L\alpha}$  with Mach number is given in figure 6 for the various wings investigated. Figure 7 presents the flap-effectiveness parameter  $\alpha_\delta$  as a function of flap-chord ratio at all the test Mach numbers. A comparison of  $\alpha_\delta$  with theory is shown in figure 8.

The variation of the lift-curve slope with aspect ratio at  $M = 0.4$  (fig. 5) shows exceptional agreement with the theory of reference 8. The variation of  $C_{L\alpha}$  with  $M$  (fig. 6) is presented to give a more complete meaning to the values of  $\alpha_\delta$  presented.

The variation of the flap-effectiveness parameter  $\alpha_\delta$  with flap-chord ratio (fig. 7) is presented in order to eliminate the necessity of plotting all the lift data to see the trends with Mach number and aspect ratio. The values of  $\alpha_\delta$  plotted in figure 7 were obtained by dividing the change in angle of zero lift from the plain wing to the flapped wing by the flap deflection and therefore represent the value of  $\alpha_\delta$  at  $C_L = 0$ . This method is somewhat more accurate than using  $C_{L\delta}/C_{L\alpha}$  and except where there are nonlinearities in the lift curve give the same value. A comparison of the variation of  $\alpha_\delta$  with aspect ratio with the subsonic (ref. 5) and supersonic (ref. 6) theories (fig. 8) show that at  $M = 0.4$  and 1.1 the theories predict the variation quite satisfactorily but not the magnitude. At a Mach number of 0.9 neither theory gives either a satisfactory variation or the correct

magnitude. The disagreement between theory and experiment in magnitude is typical and results to some extent from the thickened boundary layer at the trailing edge. The magnitude at  $M = 0.4$  agrees quite well with other published data, for example reference 9. These results indicate that both in the subsonic- and supersonic-speed ranges the available theories can be used to obtain the variation of  $a_\delta$  with  $A$ , but in the transonic range the variation must be obtained from experimental studies.

It should be pointed out that these data are for only one value of  $\delta$ , and, although they are useful in determining the effects of the several variables, they are not necessarily applicable to the design of a control surface that uses small deflections in the transonic-speed range (see ref. 10).

Langley Aeronautical Laboratory,  
National Advisory Committee for Aeronautics,  
Langley Field, Va., May 4, 1956.

## REFERENCES

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TABLE 1.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 1 MODEL

$$t_{\text{plot}} = 0.04 \quad \frac{c_f}{c_i} = \text{NONE}$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-4253					-10	-4323	.0825	.0001	-0.0490	.0150
-7	-2982					-7	-2832	.0440	-0.0081	-0.0316	.0096
-5	-2324					-5	-2025	.0276	-0.0139	-0.0222	.0071
-3	-1666					-3	-1280	.0201	-0.0138	-0.0147	.0049
-2	-1228					-2	-0783	.0196	-0.0084	-0.0094	.0033
-1	-0965					-1	-0534	.0164	-0.0065	-0.0057	.0018
0	-0702					0	-0199	.0164	-0.0009	-0.0026	.0018
1	-0351					1	.0087	.0159	.0023	.0015	.0024
2	-0132					2	.0460	.0196	.0090	.0045	.0025
3	.0088					3	.0832	.0214	.0119	.0094	.0042
5	.0570					5	.1665	.0306	.0122	.0170	.0071
7	.1272					7	.2509	.0477	.0142	.0275	.0119
10	.2280					10	.3938	.0855	.0069	.0448	.0193
15	.4297					15	.6634	.1893	-0.0357	.0765	.0354
20	.6884					20	.8870	.3421	.1009	.1044	.0529
25	.8419					25	.9689	.4765	-0.1365	.1123	.0672
$M = 0.60$											
-10	-4089	.0705	-0.0178	-0.0383	.0193	$M = 1.00$					
-7	-2915	.0461	-0.0135	-0.0310	.0120	-10	-4396	.0901	.0047	-0.0505	.0121
-5	-2132	.0374	-0.0183	-0.0225	.0082	-7	-2887	.0468	-0.0098	-0.0332	.0074
-3	-1414	.0331	-0.0119	-0.0152	.0063	-5	-1937	.0292	-0.0126	-0.0223	.0050
-2	-1044	.0322	-0.0134	-0.0112	.0054	-3	-1164	.0216	-0.0147	-0.0137	.0042
-1	-0740	.0322	-0.0116	-0.0072	.0044	-2	-0748	.0204	-0.0114	-0.0090	.0031
0	-0435	.0365	-0.0048	-0.0033	.0026	-1	-0.392	.0192	-0.0059	-0.047	.0023
1	-0174	.0374	-0.0002	.0000	.0035	0	.0154	.0188	-0.0017	-0.0025	.0012
2	.0044	.0374	-0.0013	.0033	.0038	1	.0143	.0188	.0013	.0011	.0014
3	.0392	.0426	.0035	.0079	.0051	2	.0558	.0192	.0115	.0051	.0014
5	.1022	.0579	.0021	.0165	.0054	3	.0855	.0221	.0165	.0090	.0031
7	.1784	.0748	.0106	.0257	.0089	5	.1711	.0321	.0172	.0180	.0050
10	.3197	.1122	.0296	.0416	.0016	7	.2756	.0513	.0092	.0296	.0117
15	.5068	.2162	-0.0173	.0673	.0294	10	.4301	.0946	-0.0071	.0490	.0173
20	.7830	.3689	-0.0825	.0983	.0506	15	.7105	.2074	-0.0575	.0829	.0337
25	.9027	.5029	-0.1325	.1089	.0699	20	.9362	.3600	-0.1668	.1103	.0571
$M = 0.80$											
-10	-4059	.0717	-0.0080	-0.0457	.0150	$M = 1.05$					
-7	-2848	.0398	-0.0130	-0.0304	.0092	-10	-4471	.0894	.0143	-0.0511	.0081
-5	-2066	.0289	-0.0136	-0.0215	.0058	-7	-2878	.0480	-0.0064	-0.0331	.0055
-3	-1299	.0254	-0.0097	-0.0134	.0039	-5	-1854	.0291	.0168	.0214	.0023
-2	-0826	.0239	-0.0045	-0.0099	.0030	-3	-1229	.0223	.0169	.0138	.0002
-1	-0517	.0233	-0.0001	-0.0067	.0030	-2	-0.751	.0207	-0.0103	-0.0086	-0.0020
0	-0236	.0239	.0010	-0.0027	.0028	-1	-0.466	.0212	-0.0078	-0.0052	-0.0022
1	.0030	.0233	.0017	.0013	.0028	0	.0148	.0191	-0.0023	-0.0010	-0.0017
2	.0280	.0254	.0084	.0045	.0032	1	.0159	.0207	.0001	.0034	-0.0053
3	.0635	.0289	.0083	.0080	.0049	2	.0592	.0218	.0077	.0066	-0.0020
5	.1417	.0398	.0120	.0166	.0049	3	.0933	.0234	.0110	.0104	-0.0007
7	.2022	.0543	.0147	.0255	.0082	5	.1752	.0346	.0125	.0193	.0020
10	.3335	.0921	.0172	.0403	.0172	7	.2730	.0526	.0039	.0304	.0061
15	.5564	.1966	-0.0176	.0694	.0311	10	.4380	.0951	-0.0181	.0493	.0144
20	.8235	.3338	-0.0962	.0954	.0493	15	.7167	.2155	-0.0655	.0846	.0293
25	.8973	.4501	-0.1333	.1052	.0607	20	.9488	.3770	-0.1176	.1149	.0518
$M = 0.90$											
-10	-4232	.0781	-0.0047	-0.0474	.0159	$M = 1.10$					
-7	-2839	.0417	-0.0119	-0.0316	.0104	-10	-4467	.0823	.0158	-0.0515	.0053
-5	-1901	.0268	-0.0068	-0.0217	.0078	-7	-2934	.0442	-0.0052	-0.0339	.0008
-3	-1276	.0193	-0.0118	-0.0134	.0055	-5	-1960	.0274	-0.0133	-0.0219	-0.0011
-2	-0755	.0185	-0.0076	-0.0091	.0047	-3	-1204	.0195	-0.0132	-0.0133	-0.0025
-1	-0560	.0167	-0.0065	-0.0059	.0034	-2	-0.766	.0173	-0.0098	-0.0083	-0.0027
0	-0234	.0172	-0.0010	-0.0020	.0025	-1	-0.471	.0155	-0.0054	-0.0050	-0.0027
1	.0026	.0159	.0004	.0020	.0013	0	.0120	.0151	-0.0031	-0.0017	-0.0030
2	.0378	.0193	.0092	.0051	.0017	1	.0131	.0145	-0.0003	.0027	-0.0035
3	.0677	.0198	.0062	.0091	.0034	2	.0602	.0204	.0099	.0067	-0.0022
5	.1589	.0320	.0113	.0174	.0070	3	.0876	.0232	.0121	.0100	-0.0033
7	.2370	.0461	.0165	.0269	.0125	5	.1741	.0344	.0108	.0199	-0.0014
10	.3737	.0833	.0096	.0435	.0201	7	.2628	.0539	.0028	.0299	.0014
15	.6368	.1922	-0.0250	.0755	.0337	10	.4204	.1023	-0.0183	.0482	.0083
20	.8516	.3331	-0.1014	.0976	.0494	15	.7051	.2277	.0675	.0830	.0231
25	.9298	.4560	-0.1371	.1086	.0632	20	.9416	.3931	-0.1206	.1156	.0452
$M = 1.00$											
-10	-4323	.0825	.0001	-0.0490	.0150	$M = 1.20$					
-7	-2832	.0440	-0.0081	-0.0316	.0096	-10	-4467	.0823	.0158	-0.0515	.0053
-5	-2025	.0276	-0.0139	-0.0222	.0071	-7	-2934	.0442	-0.0052	-0.0339	.0008
-3	-1280	.0201	-0.0138	-0.0147	.0049	-5	-1960	.0274	-0.0133	-0.0219	-0.0011
-2	-0783	.0196	-0.0084	-0.0094	.0033	-3	-1204	.0195	-0.0132	-0.0133	-0.0025
-1	-0534	.0164	-0.0065	-0.0057	.0018	-2	-0.766	.0173	-0.0098	-0.0083	-0.0027
0	-0199	.0164	-0.0009	-0.0026	.0018	-1	-0.471	.0155	-0.0031	-0.0017	-0.0030
1	.0087	.0159	.0023	.0015	.0024						
2	.0460	.0196	.0090	.0045	.0025						
3	.0832	.0214	.0119	.0094	.0042						
5	.1665	.0306	.0122	.0170	.0071						
7	.2509	.0477	.0142	.0275	.0119						
10	.3938	.0855	.0069	.0448	.0193						
15	.6634	.1893	-0.0357	.0765	.0337						
20	.8870	.3421	.1009	.1044	.0529						
25	.9689	.4765	-0.1365	.1123	.0672						

TABLE 1.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 1 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{c_f}{c} = 0.10$$

$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$	$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$
M = 0.40						M = 0.95					
-10	-0.2981					-10	-0.3042	.0507	-0.0513	-0.0354	.0200
-7	-0.2104					-7	-0.1751	.0268	-0.0705	-0.0181	.0171
-5	-0.1227					-5	-0.0993	.0209	-0.0713	-0.0075	.0144
-3	-0.0526					-3	0.0000	.0176	-0.0678	.0026	.0121
-2	-0.0044					-2	0.0335	-0.0415	-0.0691	.0072	.0126
-1	0.0307					-1	0.0807	.0226	-0.0667	.0117	.0123
0	0.0570					0	0.1056	.0268	-0.0662	.0158	.0119
1	0.0745					1	0.1465	.0318	-0.0611	.0200	.0128
2	0.1008					2	0.1888	.0373	-0.0580	.0233	.0137
3	0.1096					3	0.2173	.0390	-0.0619	.0271	.0150
5	0.1622					5	0.3080	.0586	-0.0598	.0377	.0202
7	0.2411					7	0.3850	.0849	-0.0590	.0478	.0269
10	0.3507					10	0.5377	.1356	-0.0692	.0667	.0350
15	0.5567					15	0.8072	.2620	-0.1072	.1006	.0523
20	0.8109					20	1.0059	.4262	-0.1552	.1266	.0733
25	0.9863					25	1.0853	.5764	-0.1910	.1345	.0881
M = 0.60						M = 1.00					
-10	-0.2610	.0344	-0.0732	-0.0271	.0202	-10	-0.3302	.0637	-0.0394	-0.0371	.0190
-7	-0.1631	.0278	-0.0826	-0.0152	.0148	-7	-0.1782	.0373	-0.0592	-0.0191	.0155
-5	-0.0696	.0309	-0.0849	-0.0059	.0127	-5	-0.0772	.0228	-0.0670	-0.0065	.0131
-3	-0.0152	.0331	-0.0741	.0013	.0117	-3	-0.0024	.0200	-0.0767	.0025	.0119
-2	0.0196	.0309	-0.0922	.0066	.0098	-2	0.0416	.0221	-0.0712	.0072	.0112
-1	0.0305	.0344	-0.0813	.0099	.0117	-1	0.0772	.0257	-0.0674	.0112	.0107
0	0.0631	.0183	-0.0795	.0145	.0117	0	0.1069	.0285	-0.0675	.0151	.0105
1	0.0979	.0492	-0.0685	.0178	.0117	1	0.1413	.0292	-0.0620	.0187	.0110
2	0.1196	.0522	-0.0760	.0225	.0139	2	0.1805	.0373	-0.0730	.0220	.0124
3	0.1588	.0631	-0.0651	.0257	.0145	3	0.2138	.0409	-0.0558	.0263	.0138
5	0.2284	.0792	-0.0664	.0376	.0174	5	0.2945	.0577	-0.0566	.0360	.0188
7	0.2958	.1057	-0.0620	.0462	.0221	7	0.3991	.0853	-0.0625	.0486	.0255
10	0.4350	.1649	-0.0607	.0640	.0326	10	0.5570	.1425	-0.0845	.0681	.0333
15	0.6634	.2801	-0.0917	.0937	.0553	15	.8314	.2710	-0.1285	.1024	.0506
20	0.9179	.4750	-0.1395	.1234	.0791	20	1.0452	.4404	-0.1705	.1323	.0734
25	1.0636	.6290	-0.1787	.1379	.1063	25	1.1924	.6285	-0.2157	.1503	.0980
M = 0.80						M = 1.05					
-10	-0.2775	.0413	-0.0711	-0.0304	.0204	-10	-0.3470	.1297	-0.0298	-0.0400	.0132
-7	-0.1594	.0283	-0.0784	-0.0152	.0156	-7	-0.1843	.0425	-0.0571	-0.0207	.0075
-5	-0.0841	.0248	-0.0749	-0.0058	.0120	-5	-0.0853	.0314	-0.0681	-0.0083	.0078
-3	-0.0074	.0248	-0.0665	.0031	.0110	-3	-0.0205	.0257	-0.0705	.0010	.0058
-2	0.0266	.0248	-0.0692	.0076	.0110	-2	0.0364	.0273	-0.0689	.0062	.0058
-1	0.0620	.0283	-0.0676	.0112	.0110	-1	0.0705	.0291	-0.0666	.0104	.0058
0	0.1107	.0354	-0.0546	.0143	.0110	0	0.0933	.0303	-0.0645	.0138	.0059
1	0.1210	.0354	-0.0647	.0179	.0118	1	0.1160	.0319	-0.0655	.0179	.0046
2	0.1505	.0413	-0.0563	.0220	.0133	2	0.1729	.0403	-0.0555	.0214	.0061
3	0.1845	.0463	-0.0593	.0269	.0148	3	0.2036	.0441	-0.0571	.0255	.0102
5	0.2686	.0646	-0.0499	.0358	.0180	5	0.2867	.0637	-0.0577	.0349	.0122
7	0.3321	.0865	-0.0531	.0457	.0234	7	0.3834	.0890	-0.0676	.0466	.0189
10	0.4649	.1393	-0.0548	.0855	.0352	10	0.5472	.1422	-0.0877	.0663	.0256
15	0.7010	.2642	-0.0901	.0922	.0515	15	.8077	.2735	-0.1344	.1008	.0431
20	0.9475	.4327	-0.1592	.1182	.0719	20	1.0330	.4575	-0.1770	.1325	.0612
25	1.0331	.5576	-0.1796	.1258	.0886	25	1.1922	.6534	-0.2215	.1540	.0853
M = 0.90						M = 1.10					
-10	-0.2851	.0443	-0.0597	-0.0328	.0218	-10	-0.2766	.0749	-0.0220	-0.0294	.0021
-7	-0.1627	.0250	-0.0769	-0.0158	.0172	-7	-0.1470	.0760	-0.0510	-0.0126	.0035
-5	-0.0885	.0172	-0.0773	-0.0055	.0140	-5	-0.0545	.0440	-0.0625	-0.0010	.0055
-3	-0.0130	.0206	-0.0728	.0036	.0121	-3	0.0033	.0407	-0.0728	.0083	.0063
-2	0.0417	.0185	-0.0685	.0079	.0121	-2	0.0534	.0418	-0.0674	.0126	.0073
-1	0.0755	.0219	-0.0669	.0118	.0123	-1	0.0926	.0440	-0.0629	.0162	.0078
0	0.1146	.0250	-0.0661	.0158	.0123	0	0.1056	.0451	-0.0668	.0201	.0065
1	0.1406	.0281	-0.0643	.0198	.0125	1	0.1394	.0481	-0.0624	.0238	.0066
2	0.1835	.0364	-0.0562	.0233	.0142	2	0.1906	.0547	-0.0529	.0271	.0084
3	0.2109	.0396	-0.0581	.0272	.0146	3	0.2330	.0599	-0.0502	.0311	.0117
5	0.2890	.0583	-0.0563	.0364	.0193	5	0.3071	.0815	-0.0536	.0403	.0133
7	0.3749	.0820	-0.0545	.0470	.0257	7	0.3920	.1093	-0.0623	.0506	.0169
10	0.5116	.1356	-0.0584	.0648	.0350	10	0.5336	.1703	-0.0848	.0690	.0245
15	0.7810	.2541	-0.0937	.0956	.0520	15	.7950	.3043	-0.1237	.1021	.0378
20	0.9893	.4251	-0.1635	.1201	.0706	20	1.0128	.4735	-0.1652	.1335	.0560
25	1.0284	.5493	-0.1851	.1272	.0849	25	1.1631	.6556	-0.2090	.1559	.0769
-10	-0.2973					-10	-0.2973	.0686	-0.0283	-0.0317	.0082
-2	.0512					-2	.0512	.0331	-0.0685	.0122	.0087

TABLE 1.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 1 MODEL - Continued

$$\frac{L}{c} = 0.04 \quad \frac{C_f}{c} = 0.20$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.2322					-10	-0.2334	.0519	-0.0758	-0.0264	.0078
-7	-0.1227					-7	-0.0881	.0281	-0.0888	-0.0109	.0056
-5	-0.0570					-5	-0.0062	.0214	-0.0958	0.0000	.0047
-3	-0.0000					-3	.0646	.0226	-0.0947	.0083	.0048
-2	.0526					-2	.1179	.0276	-0.0945	.0132	.0045
-1	.0833					-1	.1477	.0293	-0.0934	.0169	.0051
0	.1183					0	.1825	.0348	-0.0947	.0211	.0054
1	.1402					1	.2135	.0377	-0.0964	.0252	.0056
2	.1709					2	.2570	.0469	-0.0886	.0282	.0067
3	.1972					3	.2880	.0524	-0.0880	.0320	.0092
5	.2410					5	.3687	.0745	-0.0718	.0411	.0137
7	.3330					7	.4581	.1008	-0.0886	.0508	.0191
10	.4250					10	.6108	.1587	-0.0989	.0697	.0274
15	.6179					15	.8864	.2930	-0.1318	.1017	.0457
20	.8238					20	1.0726	.4531	-0.1654	.1250	.0664
25	1.0254					25	1.0949	.5837	-0.1910	.1642	.0778
$M = 0.60$											
$M = 1.00$											
-10	-0.2195	.0265	-0.0800	-0.0264	.0110	-10	-0.2576	.0641	-0.0733	-0.0299	.0081
-7	-0.1022	.0170	-0.0811	-0.0132	.0070	-7	-0.0962	.0321	-0.0894	-0.0119	.0047
-5	-0.0326	.0161	-0.0787	-0.0040	.0057	-5	-0.0024	.0264	-0.1023	0.0000	.0045
-3	.0413	.0248	-0.0760	.0040	.0054	-3	.0712	.0264	-0.1012	.0083	.0045
-2	.0696	.0265	-0.0784	.0086	.0051	-2	.1175	.0285	-0.1046	.0133	.0045
-1	.1025	.0322	-0.0734	.0112	.0063	-1	.1579	.0321	-0.1024	.0173	.0048
0	.1282	.0322	-0.0801	.0165	.0044	0	.1864	.0351	-0.0991	.0213	.0053
1	.1717	.0387	-0.0731	.0191	.0066	1	.2220	.0408	-0.0973	.0245	.0064
2	.1978	.0448	-0.0798	.0231	.0089	2	.2612	.0480	-0.0942	.0281	.0079
3	.2282	.0535	-0.0709	.0271	.0101	3	.2873	.0525	-0.0905	.0313	.0095
5	.2978	.0748	-0.0757	.0362	.0127	5	.3740	.0748	-0.0908	.0403	.0134
7	.3739	.1039	-0.0707	.0462	.0177	7	.4713	.1021	-0.0971	.0515	.0183
10	.4999	.1582	-0.0668	.0620	.0272	10	.6387	.1593	-0.1182	.0695	.0262
15	.7042	.2756	-0.0868	.0877	.0449	15	.8833	.2978	-0.1528	.1009	.0437
20	.9542	.4512	-0.1383	.1134	.0711	20	1.0852	.4694	-0.1899	.1272	.0659
25	1.0998	.6156	-0.1638	.1247	.0954	25	1.2063	.6492	-0.2145	.1423	.0918
$M = 0.80$											
$M = 1.05$											
-10	-0.2110	.0363	-0.0750	-0.0237	.0099	-10	-0.2651	.0698	-0.0701	-0.0297	.0061
-7	-0.0989	.0239	-0.0817	-0.0112	.0058	-7	-0.1172	.0437	-0.0914	-0.0138	.0010
-5	-0.0236	.0195	-0.0815	-0.0009	.0045	-5	-0.0148	.0337	-0.1017	-0.0017	.0003
-3	.0546	.0218	-0.0835	.0076	.0043	-3	.0592	.0307	-0.1056	.0076	.0020
-2	.0959	.0254	-0.0747	.0112	.0045	-2	.1081	.0307	-0.1037	.0138	.0018
-1	.1181	.0275	-0.0824	.0139	.0045	-1	.1411	.0337	-0.1027	.0162	.0023
0	.1609	.0328	-0.0728	.0179	.0052	0	.1729	.0364	-0.1014	.0200	.0026
1	.1830	.0363	-0.0785	.0224	.0064	1	.2036	.0411	-0.1012	.0235	.0035
2	.2214	.0422	-0.0712	.0260	.0077	2	.2469	.0476	-0.0941	.0269	.0053
3	.2538	.0493	-0.0762	.0300	.0077	3	.2810	.0514	-0.0928	.0311	.0069
5	.3350	.0697	-0.0734	.0385	.0120	5	.3640	.0739	-0.0928	.0390	.0117
7	.4118	.0945	-0.0773	.0479	.0172	7	.4539	.1019	-0.0984	.0501	.0150
10	.5387	.1508	-0.0738	.0636	.0251	10	.6075	.1568	-0.1170	.0673	.0236
15	.7645	.2671	-0.1019	.0896	.0421	15	.8578	.2933	-0.1553	.0991	.0408
20	.9918	.4383	-0.1603	.1142	.0524	20	1.0739	.4721	-0.1952	.1267	.0628
25	1.0390	.5661	-0.1818	.1191	.0753	25	1.2241	.6714	-0.2262	.1453	.0880
$M = 0.90$											
$M = 1.10$											
-10	-0.2265	.0469	-0.0790	-0.0245	.0076	-10	-0.2639	.0613	-0.0656	-0.0299	.0008
-7	-0.0924	.0255	-0.0878	-0.0095	.0049	-7	-0.1380	.0377	-0.0858	-0.0140	.0000
-5	-0.0143	.0180	-0.0897	.0000	.0042	-5	-0.0252	.0269	-0.0946	-0.0017	-.0013
-3	.0677	.0206	-0.0939	.0087	.0040	-3	.0515	.0258	-0.1033	.0076	-.0006
-2	.1159	.0242	-0.0881	.0130	.0042	-2	.1062	.0280	-0.1035	.0120	-.0000
-1	.1393	.0268	-0.0869	.0158	.0047	-1	.1358	.0296	-0.1019	.0156	.0010
0	.1718	.0320	-0.0844	.0198	.0049	0	.1719	.0339	-0.1011	.0199	.0011
1	.2070	.0364	-0.0886	.0233	.0066	1	.1993	.0420	-0.1014	.0236	.0024
2	.2473	.0448	-0.0784	.0265	.0077	2	.2452	.0447	-0.0935	.0266	.0035
3	.2695	.0500	-0.0787	.0308	.0097	3	.2759	.0523	-0.0914	.0309	.0053
5	.3567	.0755	-0.0814	.0395	.0138	5	.3580	.0742	-0.0923	.0399	.0091
7	.4517	.0992	-0.0845	.0505	.0200	7	.4566	.1040	-0.0959	.0502	.0139
10	.5884	.1523	-0.0902	.0679	.0286	10	.6022	.1627	-0.1184	.0668	.0217
15	.8305	.2786	-0.1098	.0968	.0463	15	.8430	.3070	-0.1574	.0973	.0382
20	1.0284	.4418	-0.1678	.1177	.0638	20	1.0445	.4771	-0.1929	.1232	.0582
25	1.0596	.5722	-0.1846	.1224	.0740	25	1.2044	.6676	-0.2289	.1432	.0802

TABLE 1.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 1 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{C_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
M = 0.40											
M = 0.95											
-10	-1.1532					-10	-1.1588	.0514	-0.0805	-0.0166	.0099
-7	-0.0350					-7	-0.0174	.0323	-0.0908	-0.0004	.0072
-5	.0569					-5	.0608	.0275	-0.0907	-0.0087	.0060
-3	.1094					-3	.1241	.0288	-0.0961	.0173	.0056
-2	.1313					-2	.1799	.0335	-0.0968	.0222	.0060
-1	.1663					-1	.2209	.0377	-0.0955	.0260	.0069
0	.2101					0	.2482	.0439	-0.0981	.0309	.0072
1	.2320					1	.2829	.0506	-0.0966	.0354	.0081
2	.2408					2	.3227	.0598	-0.0914	.0392	.0097
3	.2714					3	.3636	.0683	-0.0968	.0441	.0123
5	.3590					5	.4492	.0928	-0.0950	.0535	.0171
7	.4071					7	.5460	.1251	-0.1015	.0652	.0229
10	.5340					10	.7074	.1857	-0.1136	.0843	.0321
15	.7004					15	.9481	.1648	-0.1349	.1163	.0501
20	.9018					20	1.1343	.2514	-0.1592	.1416	.0711
25	1.1162					25	1.1045	.3040	-0.1574	.1371	.0785
M = 0.60											
M = 1.00											
-10	-1.1412	.0460	-0.0725	-0.0172	.0139	-10	-1.1840	.0608	-0.0841	-0.0184	.0112
-7	-0.0326	.0374	-0.0713	-0.0040	.0089	-7	-0.0380	.0404	-0.0994	-0.0015	.0076
-5	.0365	.0733	.0046	.0079		-5	.0629	.0321	-1.1042	.0105	.0064
-3	.1086	.0426	-0.0715	.0125	.0066	-3	.1365	.0328	-1.1053	.0187	.0064
-2	.1412	.0482	-0.0690	.0165	.0070	-2	.1959	.0368	-1.126	.0245	.0069
-1	.1716	.0513	-0.0696	.0204	.0070	-1	.2280	.0420	-1.076	.0277	.0072
0	.2064	.0586	-0.0648	.0250	.0070	0	.2576	.0480	-1.1114	.0313	.0074
1	.2324	.0643	-0.0673	.0290	.0073	1	.2873	.0537	-1.1114	.0357	.0078
2	.2607	.0717	-0.0692	.0323	.0098	2	.3384	.0632	-1.1022	.0393	.0100
3	.3041	.0799	-0.0649	.0376	.0120	3	.3681	.0712	-1.1016	.0436	.0123
5	.3910	.1047	-0.0647	.0488	.0152	5	.4630	.0976	-1.1043	.0537	.0164
7	.4605	.1334	-0.0643	.0587	.0205	7	.5580	.1285	-1.1112	.0645	.0214
10	.5930	.1925	.0630	.0778	.0329	10	.7124	.1897	-1.1294	.0832	.0293
15	.7820	.3163	-0.0856	.1015	.0493	15	.9617	.3410	-1.1674	.1167	.0480
20	1.0122	.5022	-0.1467	.1278	.0752	20	1.1516	.5162	-2.014	.1427	.0708
25	1.1515	.6729	-0.1728	.1417	.1017	25	1.2205	.6703	-2.111	.1509	.0880
M = 0.80											
M = 1.05											
-10	-1.1327	.0428	-0.0715	-0.0174	.0122	-10	-1.1933	.0643	-0.0848	-0.0186	.0083
-7	-0.0221	.0304	-0.0760	-0.0036	.0079	-7	-0.0398	.0418	-1.1036	-0.0021	.0033
-5	.0310	.0274	-0.0813	.0049	.0069	-5	.0568	.0337	-1.1115	.0093	.0040
-3	.1062	.0327	-0.0795	.0130	.0062	-3	.1364	.0337	-1.1163	.0186	.0049
-2	.1549	.0363	-0.0759	.0174	.0069	-2	.1853	.0346	-1.1183	.0228	.0046
-1	.1844	.0407	-0.0764	.0210	.0069	-1	.2217	.0391	-1.1158	.0265	.0049
0	.2271	.0472	-0.0723	.0251	.0071	0	.2524	.0448	-1.1149	.0307	.0040
1	.2478	.0522	-0.0784	.0295	.0084	1	.2853	.0514	-1.1130	.0341	.0046
2	.2832	.0587	-0.0733	.0336	.0103	2	.3308	.0587	-1.1067	.0376	.0073
3	.3171	.0675	-0.0780	.0381	.0122	3	.3638	.0682	-1.1080	.0428	.0099
5	.4071	.0920	-0.0699	.0474	.0161	5	.4491	.0928	-1.0885	.0521	.0142
7	.4867	.1212	-0.0746	.0577	.0229	7	.5559	.1257	-1.1137	.0649	.0182
10	.6150	.1799	-0.0821	.0756	.0311	10	.6980	.1844	-1.1320	.0821	.0248
15	.8407	.3047	-1.003	.1025	.0489	15	.9550	.3354	-1.1705	.1152	.0390
20	1.0472	.4787	-1.1621	.1240	.0680	20	1.1414	.5166	-2.061	.1418	.0611
25	1.0767	.6079	-1.1797	.1293	.0815	25	1.2778	.7201	-2.376	.1618	.0843
M = 0.90											
M = 1.10											
-10	-1.1470	.0447	-0.0828	-0.0150	.0110	-10	-1.1959	.0571	-0.0869	-0.0196	.0029
-7	-0.0325	.0302	-0.0910	-0.0016	.0079	-7	-0.0438	.0365	-1.1018	-0.0027	.0032
-5	.0624	.0268	-0.0851	.0083	.0059	-5	.0580	.0306	-1.1105	.0047	.0024
-3	.1210	.0289	-0.0866	.0170	.0053	-3	.1335	.0306	-1.1161	.0179	.0019
-2	.1756	.0320	-0.0903	.0213	.0055	-2	.1783	.0344	-1.1178	.0222	.0016
-1	.1977	.0359	-0.0870	.0253	.0057	-1	.2188	.0387	-1.1150	.0262	.0011
0	.2368	.0421	-0.0872	.0292	.0057	0	.2440	.0442	-1.1170	.0295	.0003
1	.2563	.0468	-0.0912	.0336	.0053	1	.2681	.0484	-1.1180	.0329	.0008
2	.3125	.0575	-0.0803	.0379	.0072	2	.3250	.0591	-1.1074	.0365	.0027
3	.3486	.0653	-0.0843	.0430	.0091	3	.3600	.0689	-1.1083	.0412	.0045
5	.4384	.0908	-0.0874	.0529	.0146	5	.4486	.0941	-1.1108	.0505	.0094
7	.5268	.1197	-0.0875	.0628	.0219	7	.5328	.1265	-1.1149	.0611	.0131
10	.6764	.1811	-0.0935	.0813	.0295	10	.6893	.1910	-1.1341	.0794	.0173
15	.9184	.3166	-1.1235	.1121	.0484	15	.9256	.3497	-1.1686	.1112	.0289
20	1.0797	.4798	-1.1716	.1318	.0671	20	1.1160	.5381	-2.144	.1408	.0490
25	1.0849	.5950	-1.1843	.1322	.0775	25	1.2386	.7318	-2.378	.1590	.0735

TABLE 1.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 1 MODEL - Concluded

$$\frac{t}{c} = 0.04 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.45$											
-10	-0.0306					-10	-1.291	.0569	-0.0972	-0.0124	.0124
-7	-0.0569					-7	.0000	.0415	-1.003	.0030	.0097
-5	.0263					-5	.0906	.0397	-1.033	.0124	.0090
-3	.1007					-3	.1564	.0439	-1.047	.0203	.0092
-2	.1226					-2	.2023	.0482	-1.009	.0249	.0097
-1	.1488					-1	.2334	.0531	-1.003	.0290	.0103
0	.1882					0	.2706	.0611	-1.030	.0335	.0106
1	.2320					1	.3017	.0690	-1.076	.0384	.0112
2	.2320					2	.3563	.0787	-1.017	.0430	.0142
3	.2670					3	.3898	.0886	-1.029	.0478	.0162
5	.3327					5	.4779	.1159	-1.036	.0588	.0222
7	.4640					7	.5710	.0886	-1.093	.0708	.0276
10	.5384					10	.7275	.2105	-1.240	.0896	.0359
15	.6916					15	.9857	.3640	-1.562	.1231	.0558
20	.9061					20	1.1545	.5373	-1.870	.1473	.0742
25	1.0637					25	1.1123	.6349	-1.863	.1393	.0821
$M = 0.60$											
$M = 1.00$											
-10	-0.0978	.0074	.0592	-0.0112	.0117	-10	-1.484	.0672	-1.011	-0.0144	.0114
-7	-0.0261	.0235	.0836	.0000	.0092	-7	-0.0024	.0449	-1.122	.0029	.0093
-5	.0304	.0448	.0720	.0086	.0114	-5	.0926	.0415	-1.133	.0137	.0083
-3	.0869	.0535	.0738	.0158	.0114	-3	.1733	.0479	-1.191	.0223	.0076
-2	.2127	.0578	.0695	.0204	.0114	-2	.2172	.0536	-1.187	.0270	.0074
-1	.1521	.0643	.0713	.0250	.0114	-1	.2587	.0596	-1.211	.0317	.0071
0	.2086	.0491	.0581	.0284	.0114	0	.2848	.0660	-1.214	.0360	.0074
1	.2390	.0761	.0768	.0336	.0120	1	.3169	.0729	-1.231	.0407	.0081
2	.2781	.0695	.0719	.0382	.0133	2	.3691	.0828	-1.169	.0414	.0105
3	.3151	.0813	.0767	.0435	.0148	3	.4071	.0933	-1.179	.0497	.0131
5	.4085	.0960	.0737	.0547	.0193	5	.4937	.1184	-1.174	.0609	.0171
7	.4780	.1369	.0800	.0646	.0256	7	.6006	.1548	-1.261	.0731	.0226
10	.6041	.2064	.0826	.0818	.0366	10	.7525	.2217	-1.453	.0918	.0318
15	.7779	.3346	.1002	.1075	.0534	15	.9875	.3760	-1.784	.0918	.0502
20	1.0321	.5267	.1648	.1371	.0802	20	1.1774	.5604	-2.138	.1512	.0756
25	1.0995	.6627	.1742	.1411	.0960	25	1.1869	.7005	-2.110	.1523	.0877
$M = 0.80$											
$M = 1.05$											
-10	-1.210	.0472	.0797	-0.0107	.0129	-10	-1.466	.0686	-1.048	-0.0145	.0071
-7	-0.758	.0398	.0653	.0013	.0097	-7	-0.0102	.0502	-1.120	.0010	.0056
-5	.0708	.0363	.0752	.0103	.0094	-5	.0863	.0436	-1.129	.0131	.0056
-3	.1372	.0437	.0797	.0179	.0099	-3	.1647	.0475	-1.263	.0214	.0049
-2	.1697	.0472	.0742	.0224	.0101	-2	.2113	.0532	-1.286	.0258	.0045
-1	.1995	.0531	.0785	.0269	.0105	-1	.2511	.0570	-1.253	.0303	.0040
0	.2375	.0596	.0737	.0304	.0107	0	.2795	.0625	-1.258	.0348	.0051
1	.2641	.0676	.0850	.0358	.0116	1	.3113	.0704	-1.292	.0393	.0059
2	.3231	.0770	.0824	.0398	.0135	2	.3704	.0811	-1.218	.0434	.0079
3	.3394	.0856	.0829	.0439	.0148	3	.4033	.0911	-1.232	.0489	.0111
5	.4308	.1148	.0783	.0546	.0204	5	.4919	.1172	-1.235	.0589	.0152
7	.5061	.1487	.0815	.0649	.0255	7	.5862	.1520	-1.337	.0700	.0197
10	.6477	.2068	.0823	.0842	.0354	10	.7226	.2168	-1.489	.0889	.0266
15	.8498	.3266	.1092	.1088	.0534	15	.9634	.3776	-1.825	.1234	.0431
20	1.0298	.4904	.1614	.1276	.0716	20	1.1497	.5699	-2.160	.1493	.0651
25	1.0298	.5949	.1710	.1294	.0804	25	1.2724	.7621	-2.488	.1679	.0888
$M = 0.90$											
$M = 1.10$											
-10	-1.249	.0544	.0950	-0.0107	.0125	-10	-1.706	.0645	-1.013	-0.0166	.0003
-7	.0078	.0396	.0935	.0032	.0093	-7	-0.0219	.0516	-1.128	.0017	.0010
-5	.0820	.0385	.0926	.0115	.0091	-5	.0700	.0473	-1.177	.0120	.0006
-3	.1444	.0435	.0945	.0194	.0091	-3	.1531	.0324	-1.225	.0202	.0010
-2	.1887	.0461	.0892	.0233	.0089	-2	.1881	.0538	-1.163	.0242	.0006
-1	.2212	.0333	.0937	.0280	.0091	-1	.2188	.0602	-1.203	.0279	.0006
0	.2577	.0609	.0914	.0324	.0093	0	.2538	.0689	-1.203	.0319	.0014
1	.2876	.0697	.0973	.0367	.0095	1	.3063	.0753	-1.170	.0375	.0030
2	.3357	.0768	.0874	.0411	.0112	2	.3500	.0860	-1.179	.0408	.0046
3	.3725	.0882	.0920	.0466	.0140	3	.3806	.0969	-1.196	.0458	.0062
5	.4581	.1153	.0928	.0564	.0187	5	.4594	.1247	-1.186	.0558	.0102
7	.5504	.1460	.0946	.0691	.0240	7	.5600	.1614	-1.296	.0674	.0151
10	.6975	.2079	.1052	.0873	.0329	10	.7000	.2367	-1.512	.0859	.0200
15	.9525	.3495	.1296	.1196	.0537	15	.9450	.3981	-1.823	.1181	.0348
20	1.0775	.5093	.1779	.1354	.0702	20	1.1288	.5917	-2.199	.1454	.0564
25	1.0671	.6145	.1810	.1354	.0789	25	1.2513	.7790	-2.506	.1653	.0792

TABLE 2.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL

 $\frac{t}{c} = 0.04$        $\frac{c_f}{c} = \text{NONE}$ 

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.95$											
-10	-4.955					-10	5.973	1.097	0.248	-0.725	.0139
-7	-3.859					-7	4.396	0.571	0.026	-0.524	.0074
-5	-2.675					-5	3.278	0.281	-0.0105	-0.364	.0043
-3	-1.710					-3	2.012	0.115	-0.148	-0.226	.0024
-2	-1.184					-2	1.428	0.076	-0.089	-0.154	.0018
-1	-0.833					-1	0.820	0.040	-0.063	-0.085	.0014
0	-0.329					0	0.286	0.031	0.030	-0.026	.0009
1	.0153					1	.0224	0.019	0.078	.0038	.0007
2	.0417					2	.0944	.0052	.0176	.0109	.0012
3	.0921					3	.1503	.0076	.0208	.0177	.0023
5	.1776					5	.2869	.0232	.0191	.0330	.0045
7	.2654					7	.4023	.0473	.0087	.0480	.0078
10	.4078					10	.5787	.0992	-0.0115	.0712	.0155
15	.6095					15	.4148	.2210	.0548	.1055	.0655
20	.7366					20	.4731	.3604	-0.0204	.1168	.0496
25	.6731					25	.4731	.4581	-0.0337	.1187	.0596
$M = 0.60$											
$M = 1.00$											
-10	-5.744	.0909	.0009	-0.701	.0136	-10	-6.342	1.227	0.504	-0.766	.0155
-7	-4.069	.0481	-0.284	-0.464	.0075	-7	-4.608	.0672	0.211	-0.537	.0081
-5	-2.872	.0294	-0.027	-0.327	.0043	-5	-3.302	.0380	0.009	-0.378	.0041
-3	-1.893	.0231	-0.018	-0.211	.0028	-3	-2.043	.0190	-0.0145	-0.225	.0019
-2	-1.327	.0198	-0.0154	-0.149	.0021	-2	-1.354	.0126	-0.0120	-0.150	.0012
-1	-0.849	.0187	-0.0136	-0.111	.0019	-1	-0.784	.0102	-0.0100	-0.081	.0009
0	-0.261	.0187	-0.0046	-0.036	.0016	0	-0.214	.0072	-0.0029	-0.014	.0007
1	.0131	.0187	-0.0063	.0023	.0016	1	.0499	.0094	-0.0010	.0067	.0009
2	.0609	.0220	.0024	.0078	.0017	2	.1116	.0116	.0082	.0135	.0016
3	.1131	.0242	.0015	.0140	.0022	3	.1686	.0160	.0103	.0207	.0026
5	.2089	.0374	.0101	.0264	.0047	5	.3183	.0350	-0.0029	.0373	.0055
7	.3177	.0616	.0146	.0396	.0076	7	.4276	.0584	-0.0133	.0514	.0086
10	.4874	.1123	.0141	.0611	.0146	10	.6247	.1125	-0.0438	.0757	.0173
15	.6984	.2230	-0.0507	.0891	.0265	15	.8979	.2394	-0.0812	.1117	.0354
20	.7746	.3198	-0.0980	.0974	.0337	20	1.0713	.4007	-0.1218	.1369	.0556
25	.7485	.3884	-0.0971	.0941	.0498	25	1.1497	.5549	-0.1864	.1449	.0725
$M = 0.80$											
$M = 1.05$											
-10	-5.891	.0918	.0020	-0.692	.0131	-10	-6.553	1.306	0.689	-0.752	.0141
-7	-4.340	.0433	.0196	-0.488	.0072	-7	-4.641	.0713	.0324	-0.526	.0070
-5	-3.015	.0199	.0165	-0.336	.0041	-5	-3.390	.0433	.0161	-0.375	.0047
-3	-2.037	.0097	.0091	-0.213	.0022	-3	-2.207	.0251	.0050	-0.233	.0014
-2	-1.388	.0062	.0085	-0.146	.0015	-2	-1.502	.0188	.0025	-0.155	.0006
-1	-0.827	.0044	.0063	-0.081	.0012	-1	-0.887	.0146	-0.0017	-0.088	.0003
0	-0.251	.0037	.0018	-0.027	.0010	0	-0.205	.0117	-0.0028	-0.017	.0003
1	.0165	.0025	.0041	.0029	.0009	1	.0387	.0126	-0.0019	.0057	.0000
2	.0723	.0090	.0104	.0092	.0011	2	.1160	.0159	.0008	.0138	.0014
3	.1270	.0090	.0119	.0159	.0017	3	.1797	.0196	-0.0033	.0212	.0027
5	.2480	.0229	.0226	.0298	.0043	5	.3185	.0370	.0116	.0368	.0050
7	.3635	.0446	.0261	.0441	.0072	7	.4368	.0613	-0.0244	.0509	.0081
10	.5433	.0948	.0152	.0665	.0142	10	.6302	.1183	-0.0549	.0751	.0159
15	.6998	.1961	-0.0535	.0878	.0284	15	.9328	.2590	-0.1090	.1142	.0348
20	.7707	.2941	-0.0890	.0952	.0393	20	1.1558	.4308	-0.1492	.1432	.0577
25	.7825	.1924	-0.0751	.0974	.0488	25	1.2468	.5930	-0.1942	.1553	.0774
$M = 0.90$											
$M = 1.10$											
-10	-5.975	.1002	.0135	-0.717	.0137	-10	-6.438	1.225	0.709	-0.0070	.0113
-7	-4.543	.0512	-0.038	-0.519	.0079	-7	-4.796	.0670	.0332	-0.520	.0052
-5	-3.176	.0240	-0.015	-0.361	.0049	-5	-3.460	.0391	.0171	-0.365	.0018
-3	-2.083	.0096	.0120	-0.223	.0027	-3	-2.277	.0229	.0069	-0.228	.0000
-2	-1.419	.0044	.0103	-0.148	.0020	-2	-1.533	.0177	.0031	-0.158	-.0008
-1	-0.833	.0016	-0.0062	-0.089	.0017	-1	-0.920	.0142	.0003	-0.091	-.0012
0	-0.247	.0009	.0019	-0.024	.0013	0	-0.263	.0122	-0.0010	-0.017	-.0014
1	.0208	.0000	.0074	.0034	.0012	1	.0416	.0113	-0.0028	.0061	-.0009
2	.0794	.0033	.0144	.0099	.0013	2	.1117	.0162	-0.0008	.0133	-.0000
3	.1445	.0070	.0178	.0174	.0025	3	.1774	.0216	-0.0028	.0204	-.0010
5	.2747	.0228	.0233	.0326	.0050	5	.3066	.0382	-0.0136	.0350	-.0036
7	.4061	.0448	.0141	.0480	.0086	7	.4292	.0646	-0.0282	.0507	-.0061
10	.5702	.0937	.0013	.0693	.0156	10	.6131	.1198	-0.0593	.0728	-.0127
15	.8097	.2145	-0.0349	.1027	.0326	15	.9000	.2569	-0.1113	.1103	.0314
20	.8696	.3291	-0.1143	.1066	.0445	20	1.1430	.4362	-0.1620	.1412	.0535
25	.8722	.4225	-0.1238	.1086	.0558	25	1.8613	.6246	-0.5222	.1601	.0786

TABLE 2.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{C_f}{c} = 0.10$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.55$											
-10	-2775					-10	-4587	.0884	-0.0492	-0.0551	.0132
-7	-2032					-7	-2851	.0436	-0.0767	-0.0320	.0080
-5	-1093					-5	-1674	.0234	-0.0881	-0.0164	.0064
-3	-0109					-3	-0322	.0140	-0.0987	-0.0009	.0051
-2	.0262					-2	.0434	.0131	-0.0999	-0.0073	.0047
-1	.0699					-1	.1091	.0152	-0.1005	.0151	.0051
0	.1093					0	.1736	.0192	-0.1030	.0226	.0058
1	.1639					1	.2393	.0247	-0.0955	.0301	.0073
2	.1945					2	.3075	.0324	-0.0936	.0367	.0086
3	.2447					3	.3744	-.1116	-0.0940	.0451	.0099
5	.3496					5	.5232	.0689	-0.0988	.0630	.0135
7	.4370					7	.6447	.1015	-0.1610	.0784	.0181
10	.5856					10	.8033	.1665	-0.1186	.1004	.0272
15	.7539					15	1.0166	.3024	-0.1376	.1316	.0467
20	.8303					20	1.0414	.4280	-0.1955	.1316	.0589
25	.7473					25	1.0191	.5316	-0.2014	.1298	.0674
$M = 0.60$											
$M = 1.00$											
-10	-3578	.0256	-.0698	-.0433	.0126	-10	-5336	.1059	-0.0122	-0.0624	.0155
-7	-2169	.0124	-.1011	-.0304	.0079	-7	-3415	.0618	-0.0485	-0.0387	.0094
-5	-1063	-.1546	-.1029	-.0089	.0055	-5	-2075	.0373	-0.0669	-0.0210	.0069
-3	-.0054	.0059	-.0885	.0026	.0041	-3	-.0771	.0250	-0.0801	-0.0063	.0054
-2	.0520	.0085	-.0890	.0094	.0039	-2	.0036	.0210	-0.0905	.0036	.0049
-1	.0954	.0113	-.0879	.0150	.0039	-1	.0711	.0213	-0.0944	.0108	.0050
0	.1453	.0150	-.0825	.0206	.0043	0	.1328	.0242	-0.0948	.0180	.0053
1	.1973	.0182	-.0806	.0268	.0050	1	.2170	.0286	-0.0963	.0273	.0069
2	.2429	.0245	-.0783	.0331	.0069	2	.2846	.0373	-0.0970	.0342	.0080
3	.3014	.0362	-.0752	.0398	.0087	3	.3557	.0461	-0.1025	.0432	.0095
5	.4023	.0618	.0679	.0530	.0128	5	.5039	.0709	-0.1152	.0603	.0136
7	.5226	.0939	-.0546	.0674	.0159	7	.6450	.1079	-0.1319	.0764	.0181
10	.6874	.1572	-.0663	.0888	.0248	10	.8252	.1767	-0.1564	.1007	.0272
15	.8240	.2752	-.1178	.1089	.0206	15	1.0908	.3335	-0.1847	.1381	.0479
20	.8479	.3732	-.1483	.1102	.0262	20	1.2473	.4934	-0.2087	.1565	.0670
25	.8067	.4458	-.1459	.1063	.0307	25	1.2046	.6135	-0.2374	.1511	.0774
$M = 0.80$											
$M = 1.05$											
-10	-4020	.0558	-.0643	-.0475	.0119	-10	-5561	.1125	.0081	-.0654	.0135
-7	-2356	.0218	-.0992	-.0257	.0081	-7	-3609	.0650	-.0302	-.0405	.0071
-5	-1178	.0105	-.0959	-.0109	.0062	-5	-.2179	.0419	-.0541	-.0236	.0045
-3	-.0147	.0090	-.0883	.0016	.0045	-3	-.0942	.0291	-.0684	-.0086	.0040
-2	.0589	.0102	-.0867	.0088	.0039	-2	-.0193	.0260	-.0741	-.0004	.0032
-1	.1104	.0102	-.0854	.0151	.0047	-1	.0454	.0254	-.0752	-.0069	.0034
0	.1723	.0144	-.0818	.0220	.0053	0	.1101	.0268	-.0809	-.0148	-.0045
1	.2253	.0211	-.0857	.0290	.0066	1	.1736	.0302	-.0855	-.0224	-.0049
2	.2915	.0290	-.0821	.0353	.0082	2	.2508	.0379	-.0837	-.0293	-.0059
3	.3534	.0373	-.0753	.0430	.0094	3	.3178	.0457	-.0906	-.0374	-.0078
5	.4667	.0618	-.0654	.0578	.0120	5	.4653	.0725	-.1081	-.0549	.0120
7	.5816	.0908	-.0604	.0726	.0163	7	.6061	.1091	-.1267	.0713	.0139
10	.7539	.1543	-.0738	.0949	.0255	10	.7945	.1775	-.1587	.0961	.0226
15	.8393	.2628	-.1350	.1072	.0390	15	1.0714	.3349	-.1969	.1333	.0437
20	.8775	.3722	-.1593	.1112	.0512						
25	.8628	.4561	-.1598	.1108	.0619						
$M = 0.90$											
$M = 1.10$											
-10	-4313	.0747	-.0659	-.0516	.0122	-10	-5409	.1037	.0138	-.0635	.0092
-7	-2754	.0364	-.0814	-.0305	.0080	-7	-3693	.0591	-.0305	-.0403	.0039
-5	-1507	.0169	-.0943	-.0148	.0062	-5	-.2240	.0392	-.0485	-.0232	.0018
-3	.0260	.0105	-.0940	-.0006	.0048	-3	-.1093	.0258	-.0679	-.0091	.0007
-2	.0468	.0103	-.0935	.0073	.0046	-2	-.0240	.0241	-.0675	-.0010	.0006
-1	.1013	.0121	-.0985	.0144	.0050	-1	.0295	.0228	-.0737	-.0058	.0009
0	.1689	.0169	-.0932	.0219	.0056	0	.1005	.0263	-.0759	-.0136	.0010
1	.2338	.0249	-.0923	.0296	.0073	1	.1508	.0291	-.0819	-.0239	.0018
2	.2988	.0307	-.0871	.0363	.0085	2	.2382	.0403	-.0792	-.0280	.0035
3	.3638	.0387	-.0877	.0443	.0097	3	.3005	.0484	-.0859	-.0356	.0051
5	.5067	.0648	-.0838	.0617	.0139	5	.4349	.0747	-.1020	-.0519	.0075
7	.6301	.0972	-.0889	.0773	.0186	7	.5704	.1075	-.1219	-.0678	.0106
10	.7873	.1616	-.0995	.0985	.0266	10	.7539	.1790	-.1553	-.0928	.0182
15	.9042	.2766	-.1443	.1143	.0415	15	1.0315	.1698	-.1839	.1293	.0368
20	.9665	.4000	-.1765	.1202	.0574						
25	.9613	.4983	-.1820	.1210	.0640						

TABLE 2 -- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{c_f}{c} = 0.20$$

$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$	$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$
$M = 0.40$											
$M = 0.55$											
-10	-0.2164					-10	-0.3739	.0935	-0.0914	-0.0418	.0102
-7	-0.1027					-7	-0.1981	.0524	-0.1149	-0.0188	.0055
-5	-0.0109					-5	-0.0520	.0340	-0.1281	-0.0019	.0038
-3	.0830					-3	.0718	.0290	-0.1373	.0124	.0038
-2	.1333					-2	.1498	.0311	-0.1412	.0201	.0038
-1	.1704					-1	.2229	.0357	-0.1459	.0279	.0047
0	.2141					0	.2860	.0402	-0.1413	.0352	.0054
1	.2797					1	.3442	.0453	-0.1369	.0418	.0064
2	.3015					2	.4160	.0560	-0.1332	.0498	.0085
3	.3518					3	.4804	.0655	-0.1290	.0573	.0102
5	.4392					5	.6265	.0935	-0.1388	.0751	.0148
7	.5375					7	.7379	.1285	-0.1404	.0892	.0204
10	.6730					10	.8865	.1961	-0.1494	.1108	.0299
15	.8194					15	1.0474	.3251	-0.1555	.1349	.0475
20	.8806					20	1.0771	.4591	-0.2013	.1341	.0610
25	.7888					25	1.0474	.5565	-0.2050	.1672	.0711
$M = 0.60$											
$M = 1.00$											
-10	-0.2656	.0475	-.0836	-.0304	.0067	-10	-0.4346	.1098	-.0630	-.0490	.0121
-7	-0.1051	.0208	-.1136	-.0099	.0039	-7	-0.2534	.0690	-.0948	-.0252	.0068
-5	.0044	.0171	-.1109	.0033	.0024	-5	-0.1089	.0487	-.1172	-.0075	.0050
-3	.1065	.0197	-.1077	.0148	.0023	-3	.0403	.0385	-.1418	.0088	.0041
-2	.1604	.0223	-.1052	.0210	.0020	-2	.1220	.0375	-.1461	.0180	.0039
-1	.2070	.0267	-.1037	.0266	.0028	-1	.2013	.0391	-.1529	.0264	.0041
0	.2634	.0330	-.0910	.0326	.0035	0	.2759	.0443	-.1534	.0343	.0050
1	.3067	.0384	-.0960	.0380	.0047	1	.3386	.0501	-.1516	.0413	.0062
2	.3501	.0479	-.0951	.0436	.0056	2	.4121	.0588	-.1492	.0492	.0082
3	.4075	.0596	-.0888	.0501	.0067	3	.4772	.0696	-.1515	.0566	.0099
5	.5105	.0837	-.0774	.0633	.0103	5	.6134	.0996	-.1582	.0727	.0144
7	.6244	.1168	-.0733	.0776	.0158	7	.7341	.1359	-.1679	.0871	.0197
10	.7609	.1786	-.0752	.0970	.0253	10	.9070	.2079	-.1825	.1103	.0300
15	.8931	.2942	-.1291	.1135	.0410	15	1.1201	.3517	-.1898	.1419	.0511
20	.8953	.3891	-.1549	.1118	.0512	20	1.2930	.5253	-.2223	.1599	.0723
25	.8411	.4574	-.1489	.1085	.0585						
$M = 0.80$											
$M = 1.05$											
-10	-0.2935	.0597	-.0948	-.0325	.0064	-10	-0.4637	.1135	-.0415	-.0516	.0113
-7	-0.1155	.0285	-.1238	-.0104	.0039	-7	-0.2676	.0703	-.0832	-.0275	.0062
-5	.0096	.0196	-.1199	.0039	.0024	-5	-0.1213	.0499	-.1048	-.0098	.0041
-3	.1089	.0203	-.1192	.0156	.0024	-3	.0091	.0396	-.1240	.0052	.0035
-2	.1714	.0228	-.1207	.0227	.0027	-2	.0873	.0396	-.1306	.0134	.0034
-1	.2244	.0278	-.1186	.0285	.0034	-1	.1565	.0405	-.1354	.0208	.0038
0	.2855	.0352	-.1108	.0352	.0040	0	.2506	.0430	-.1454	.0310	.0047
1	.3406	.0391	-.1065	.0415	.0053	1	.3016	.0493	-.1439	.0370	.0054
2	.3870	.0466	-.1060	.0469	.0061	2	.3810	.0583	-.1439	.0447	.0071
3	.4532	.0578	-.0985	.0541	.0078	3	.4410	.0678	-.1505	.0525	.0092
5	.5694	.0818	-.0907	.0681	.0123	5	.5782	.0967	-.1568	.0679	.0130
7	.6930	.1154	-.0867	.0848	.0178	7	.7098	.1338	-.1689	.0834	.0182
10	.8446	.1788	-.0924	.1047	.0274	10	.8798	.2058	-.1829	.1063	.0276
15	.8917	.2837	-.1430	.1116	.0401	15	1.1565	.3646	-.2195	.1393	.0494
20	.9152	.3936	-.1628	.1138	.0514						
25	.8887	.4761	-.1605	.1116	.0622						
$M = 0.90$											
$M = 1.10$											
-10	-0.3335	.0798	-.1024	-.0374	.0083	-10	-0.4572	.1052	-.0391	-.0505	.0091
-7	-0.1778	.0437	-.1211	-.0162	.0051	-7	-0.2641	.0649	-.0773	-.0267	.0046
-5	-0.0286	.0293	-.1265	.0000	.0037	-5	-0.1255	.0469	-.0987	-.0099	.0032
-3	.0795	.0256	-.1271	.0128	.0034	-3	.0055	.0381	-.1167	.0043	.0027
-2	.1531	.0287	-.1277	.0201	.0034	-2	.0775	.0381	-.1210	.0121	.0027
-1	.2245	.0326	-.1292	.0278	.0042	-1	.1495	.0399	-.1259	.0199	.0029
0	.2790	.0374	-.1286	.0348	.0047	0	.2106	.0421	-.1323	.0265	.0038
1	.3439	.0437	-.1285	.0422	.0059	1	.2815	.0475	-.1363	.0343	.0048
2	.4065	.0549	-.1192	.0487	.0077	2	.3568	.0577	-.1361	.0419	.0064
3	.4685	.0628	-.1178	.0561	.0094	3	.4288	.0690	-.1393	.0505	.0082
5	.6152	.0899	-.1135	.0738	.0143	5	.5565	.0971	-.1527	.0654	.0120
7	.7346	.1236	-.1157	.0886	.0397	7	.6787	.1328	-.1645	.0806	.0166
10	.8591	.1864	-.1256	.1083	.0287	10	.8446	.2018	-.1787	.1010	.0253
15	.9811	.3076	-.1506	.1240	.0442	15	1.1043	.3509	-.2182	.1328	.0457
20	1.0071	.4239	-.1874	.1240	.0566						
25	.9734	.5170	-.1840	.1221	.0670						

TABLE 2.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{c_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.60$											
$M = 1.00$											
$M = 1.05$											
$M = 1.10$											
-10	-1.093					-10	-2.984	.0755	-0.0992	-0.0330	.0111
-7	-0.437					-7	-1.065	.0396	-1.230	-0.0090	.0069
-5	.0656					-5	.0452	.0284	-1.359	.0081	.0059
-3	.1486					-3	.1647	.0284	-1.448	.0220	.0060
-2	.1988					-2	.2495	.0334	-1.524	.0304	.0070
-1	.2557					-1	.3194	.0390	-1.481	.0379	.0076
0	.3125					0	.3776	.0478	-1.505	.0455	.0082
1	.3474					1	.4395	.0557	-1.443	.0524	.0097
2	.3715					2	.5076	.0657	-1.370	.0603	.0118
3	.4261					3	.5782	.0780	-1.419	.0693	.0139
5	.5048					5	.7131	.1117	-1.458	.0853	.0190
7	.5943					7	.8147	.1486	-1.498	.0999	.0238
10	.7276					10	.9360	.2149	-1.514	.1187	.0285
15	.8347					15	1.1068	.3507	-1.544	.1424	.0493
20	.9003					20	1.0895	.4701	-1.914	.1360	.0603
25	.7976					25	1.0524	.5760	-1.911	.1337	.0669
$M = 0.80$											
-10	-1.734	.0171	-0.0636	-0.0219	.0085	-10	-3.611	.0926	-0.0741	-0.0396	.0137
-7	-0.336	.0043	-0.0917	-0.0026	.0052	-7	-1.176	.0583	-1.080	-0.0158	.0090
-5	.0813	.0022	-0.0837	.0105	.0044	-5	-0.059	.0401	-1.358	.0040	.0070
-3	.1843	.0069	-0.0818	.0217	.0050	-3	.1385	.0367	-1.546	.0194	.0066
-2	.2428	.0113	-0.0779	.0281	.0057	-2	.2368	.0367	-1.584	.0291	.0070
-1	.2851	.0165	-0.0762	.0340	.0065	-1	.3043	.0439	-1.610	.0366	.0076
0	.3338	.0241	-0.0721	.0400	.0078	0	.3659	.0495	-1.602	.0439	.0082
1	.3775	.0330	-0.0755	.0464	.0093	1	.4381	.0583	-1.556	.0519	.0099
2	.4346	.0431	-0.0691	.0526	.0098	2	.5032	.0693	-1.551	.0591	.0117
3	.4823	.0544	-0.0692	.0592	.0113	3	.5684	.0810	-1.574	.0675	.0139
5	.5961	.0843	-0.0579	.0725	.0141	5	.7033	.1159	-1.659	.0833	.0189
7	.6937	.1194	-0.0549	.0858	.0199	7	.8170	.1590	-1.741	.0984	.0238
10	.8281	.1827	-0.0613	.1049	.0298	10	.9709	.2335	-1.869	.1200	.0337
15	.8931	.3018	-1.075	.1154	.0429	15	1.2172	.3937	-2.048	.1523	.0530
20	.8910	.4030	-1.293	.1131	.0528	20	1.2646	.5392	-2.135	.1595	.0726
25	.8498	.4724	-1.178	.1075	.0600	25	1.2078	.6428	-2.250	.1505	.0747
$M = 0.90$											
-10	-2.081	.0372	-0.0863	-0.0280	.0077	-10	-3.776	.0929	-0.0579	-0.0418	.0115
-7	-0.338	.0156	-1.107	-0.0029	.0047	-7	-1.1871	.0579	-1.017	-0.0177	.0076
-5	.0816	.0109	-1.063	.0105	.0039	-5	-0.0397	.0432	-1.190	-0.0005	.0063
-3	.1853	.0137	-1.046	.0228	.0044	-3	.0964	.0388	-1.377	.0145	.0059
-2	.2464	.0181	-1.048	.0306	.0052	-2	.1871	.0390	-1.480	.0240	.0061
-1	.3052	.0238	-1.002	.0358	.0065	-1	.2687	.0435	-1.528	.0323	.0064
0	.3530	.0329	-1.005	.0425	.0075	0	.3356	.0497	-1.604	.0409	.0066
1	.4045	.0391	-1.029	.0488	.0083	1	.4014	.0577	-1.633	.0488	.0073
2	.4633	.0506	-0.0877	.0551	.0095	2	.4739	.0692	-1.560	.0557	.0094
3	.5281	.0625	-0.0903	.0629	.0113	3	.5408	.0825	-1.606	.0643	.0112
5	.6413	.0915	-0.0828	.0783	.0161	5	.6689	.1154	-1.678	.0793	.0153
7	.7561	.1269	-0.0805	.0928	.0220	7	.7823	.1578	-1.763	.0874	.0194
10	.8914	.1902	-0.0862	.1102	.0313	10	.9501	.2375	-1.940	.1156	.0290
15	.9120	.3010	-1.315	.1156	.0421	15	1.2109	.4048	-2.258	.1493	.0491
20	.9267	.4066	-1.447	.1160	.0507						
25	.8973	.4933	-1.468	.1142	.0603						
$M = 1.10$											
-10	-2.530	.0599	-1.073	-0.0281	.0093	-10	-3.819	.0894	-0.0566	-0.0424	.0076
-7	-0.714	.0293	-1.256	-0.0055	.0062	-7	-1.1964	.0558	-0.986	-0.0184	.0042
-5	.0616	.0217	-1.286	.0093	.0050	-5	-0.360	.0410	-1.181	-0.0012	.0034
-3	.1719	.0239	-1.330	.0220	.0056	-3	.0873	.0384	-1.341	.0131	.0035
-2	.2530	.0278	-1.354	.0301	.0067	-2	.1746	.0410	-1.422	.0224	.0040
-1	.3049	.0322	-1.324	.0368	.0072	-1	.2401	.0445	-1.461	.0290	.0044
0	.3633	.0402	-1.292	.0440	.0081	0	.3023	.0483	-1.504	.0371	.0049
1	.4320	.0498	-1.286	.0516	.0098	1	.3721	.0574	-1.548	.0445	.0059
2	.4865	.0599	-1.174	.0579	.0114	2	.4441	.0692	-1.520	.0520	.0074
3	.5553	.0717	-1.180	.0667	.0138	3	.5074	.0821	-1.591	.0603	.0088
5	.6928	.1030	-1.198	.0834	.0187	5	.6460	.1159	-1.646	.0770	.0120
7	.7940	.1382	-1.166	.0980	.0229	7	.7573	.1600	-1.777	.0907	.0154
10	.9159	.2042	-1.246	.1157	.0318	10	.9188	.2420	-1.896	.1112	.0236
15	1.0145	.3267	-1.448	.1283	.0455	15	1.1654	.4057	-2.275	.1424	.0422
20	1.0171	.4402	-1.714	.1256	.0558						
25	.9730	.5270	-1.675	.1236	.0637						

TABLE 2 . - THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Continued

$$\frac{c}{C} = 0.04 \quad \frac{c_f}{C} = 0.040$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.55$											
-10	-0.0329					-10	-0.2217	.0856	-0.1403	-0.0287	.0055
-7	.0394					-7	-0.0286	.0550	-0.1685	-0.0055	.0015
-5	.1468					-5	.1280	.0443	-0.1779	.0119	.0007
-3	.2279					-3	.2559	.0467	-0.1815	.0261	.0006
-2	.2761					-2	.3329	.0534	-0.1837	.0346	.0012
-1	.3133					-1	.3950	.0596	-0.1788	.0414	.0023
0	.3506					0	.4509	.0681	-0.1794	.0482	.0034
1	.4009					1	.5068	.0776	-0.1778	.0560	.0048
2	.4338					2	.5839	.0861	-0.1707	.0635	.0077
3	.4776					3	.6410	.0963	-0.1675	.0714	.0103
5	.5631					5	.7528	.1267	-0.1621	.0861	.0154
7	.6748					7	.8522	.0825	-0.1638	.0999	.0213
10	.7887					10	.9590	.1137	-0.1565	.1157	.0308
15	.9158					15	1.1081	.0911	-0.1478	.1376	.0251
20	.9290					20	1.1130	.1237	-0.1772	.1327	.0312
25	.8457					25	1.0708	.1482	-0.1635	.1285	.0352
$M = 0.60$											
$M = 0.95$											
-10	-0.1250	.0267	-0.1009	-0.0244	-0.0016	-10	-0.2924	.0979	-0.1143	-0.0368	.0073
-7	.0250	.0161	-0.1220	-0.0058	-0.0043	-7	-0.0874	.0657	-0.1538	-0.0115	.0033
-5	.1337	.0176	-0.1219	.0089	-0.0032	-5	.0850	.0526	-0.1791	.0071	.0019
-3	.2261	.0252	-0.1133	.0213	-0.0008	-3	.2217	.0517	-0.1894	.0227	.0015
-2	.2816	.0294	-0.1097	.0277	.0006	-2	.3161	.0549	-0.1915	.0317	.0016
-1	.3272	.0391	-0.1047	.0345	.0017	-1	.3732	.0613	-0.1909	.0389	.0019
0	.3783	.0498	-0.1036	.0424	.0056	0	.4409	.0693	-0.1891	.0448	.0031
1	.4273	.0615	-0.0953	.0503	.0111	1	.5003	.0777	-0.1885	.0270	.0046
2	.4675	.0722	-0.0993	.0564	.0146	2	.5681	.0906	-0.1835	.0309	.0067
3	.5197	.0883	-0.0894	.0640	.0183	3	.6370	.1029	-0.1879	.0349	.0089
5	.6349	.1202	-0.0840	.0787	.0273	5	.7678	.1388	-0.1922	.0427	.0143
7	.7480	.1524	-0.0811	.0924	.0371	7	.8771	.1835	-0.2009	.0999	.0201
10	.8654	.2140	-0.0997	.1069	.0461	10	1.0292	.2630	-0.2109	.1204	.0320
15	.9197	.3251	-0.1490	.1089	.0466	15	1.2027	.3940	-0.2042	.1450	.0522
20	.9263	.4203	-0.1669	.1065	.0477	20	1.3454	.5845	-0.2359	.1630	.0743
25	.8871	.4973	-0.1633	.1023	.0561	25	1.1932	.6546	-0.2423	.1432	.0769
$M = 0.80$											
$M = 1.05$											
-10	-0.1564	.0443	-0.1045	-0.0240	.0004	-10	-0.3003	.0870	-0.1031	-0.0382	.0052
-7	.0148	.0244	-0.1360	-0.0034	-0.0019	-7	-0.1069	.0658	-0.1439	-0.0102	.0019
-5	.1299	.0236	-0.1338	.0105	.0022	-5	.0410	.0546	-0.1644	.0022	.0007
-3	.2457	.0298	-0.1295	.0238	-0.0008	-3	.1866	.0529	-0.1792	.0187	.0003
-2	.3011	.0370	-0.1280	.0310	.0003	-2	.2708	.0554	-0.1829	.0268	.0008
-1	.3490	.0453	-0.1254	.0376	.0016	-1	.3447	.0602	-0.1883	.0356	.0010
0	.4177	.0527	-0.1195	.0340	.0044	0	.4164	.0685	-0.1871	.0429	.0016
1	.4664	.0599	-0.1222	.0517	.0057	1	.4755	.0778	-0.1883	.0513	.0031
2	.5106	.0733	-0.1167	.0587	.0166	2	.5495	.0887	-0.1865	.0587	.0050
3	.5682	.0965	-0.1121	.0654	.0213	3	.6166	.1035	-0.1893	.0666	.0069
5	.6833	.1161	-0.1053	.0799	.0325	5	.7372	.1384	-0.1929	.0815	.0120
7	.8088	.1532	-0.1025	.0949	.0433	7	.8509	.1818	-0.2022	.0956	.0175
10	.9150	.2127	-0.1108	.1084	.0499	10	1.0011	.2629	-0.2182	.1174	.0289
15	.9475	.3208	-0.1606	.1093	.0383	15	1.2650	.4364	-0.2468	.1481	.0516
20	.9534	.4283	-0.1724	.1106	.0488						
25	.9239	.5109	-0.1707	.1088	.0498						
$M = 0.90$											
$M = 1.10$											
-10	-0.1810	.0689	-0.1426	-0.0250	.0035	-10	-0.3033	.0902	-0.0976	-0.0389	.0028
-7	.0020	.0417	-0.1633	-0.0028	.0005	-7	-0.1117	.0616	-0.1387	-0.0150	-0.0003
-5	.1455	.0371	-0.1628	.0125	.0001	-5	.0372	.0511	-0.1559	.0014	-0.0010
-3	.2617	.0426	-0.1670	.0269	-0.0001	-3	.1620	.0511	-0.1717	.0163	-0.0010
-2	.3386	.0481	-0.1619	.0343	.0007	-2	.2507	.0546	-0.1753	.0247	-0.0003
-1	.3946	.0557	-0.1632	.0415	.0015	-1	.3186	.0592	-0.1804	.0321	.0002
0	.4506	.0647	-0.1623	.0481	.0028	0	.3920	.0684	-0.1846	.0409	.0006
1	.4987	.0724	-0.1602	.0547	.0050	1	.4522	.0773	-0.1902	.0485	.0013
2	.5508	.0794	-0.1434	.0607	.0074	2	.5288	.0888	-0.1826	.0561	.0034
3	.6094	.0928	-0.1440	.0685	.0092	3	.5934	.1042	-0.1871	.0640	.0049
5	.7425	.1233	-0.1386	.0844	.0137	5	.7117	.1421	-0.1918	.0784	.0087
7	.8285	.1569	-0.1368	.0972	.0185	7	.8277	.1869	-0.2044	.0930	.0140
10	.9454	.2216	-0.1372	.1138	.0568	10	.9722	.2692	-0.2190	.1133	.0242
15	1.0652	.3522	-0.1554	.1300	.0890	15	1.2175	.4351	-0.2544	.1432	.0470
20	1.0443	.4675	-0.1967	.1225	.0564						
25	.9975	.5572	-0.1927	.1193	.0651						

TABLE 2-- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = \text{NONE}$$

$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$	$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$
$M = 0.40$											
-10	-0.4485					-10	-0.5376	.1123	.0288	-0.0643	-0.0133
-7	-0.3405					-7	-0.3822	.0600	-0.0008	-0.0443	-0.0068
-5	-0.2484					-5	-0.2871	.0390	-0.0098	-0.0208	-0.0031
-3	-0.1620					-3	-0.1840	.0251	-0.0063	-0.0193	-0.0002
-2	-0.1296					-2	-0.1301	.0219	-0.0049	-0.0123	.0007
-1	-0.0864					-1	-0.0666	.0193	-0.0028	-0.0054	.0022
0	-0.0378					0	-0.0143	.0155	.0028	.0011	.0022
1	.0165					1	.0523	.0181	.0042	.0089	.0026
2	.0540					2	.1078	.0219	.0098	.0146	.0026
3	.1080					3	.1649	.0257	.0112	.0212	.0024
5	.2214					5	.2680	.0406	.0112	.0346	.0033
7	.3078					7	.3822	.0647	.0008	.0292	.0009
10	.4590					10	.5376	.1132	-0.0260	.0693	-0.0044
15	.6426					15	.7962	.2309	-0.0962	.1039	-0.0201
20	.7452					20	1.0023	.3860	-0.1958	.1251	-0.0421
25	.7452					25	1.0309	.5132	-0.2379	.1282	-0.0594
$M = 0.60$											
$M = 1.00$											
-10	-0.5027	.1094	.0024	-0.0587	-0.0074	-10	-0.6184	.1346	.0760	-0.0723	-0.0163
-7	-0.3555	.0533	-0.0062	-0.0391	-0.0039	-7	-0.4467	.0808	.0410	-0.0513	-0.0090
-5	-0.2610	.0383	-0.0073	-0.0269	-0.0007	-5	-0.3252	.0553	.0242	-0.0369	-0.0044
-3	-0.1638	.0244	-0.0087	-0.0162	.0023	-3	-0.2097	.0395	.0141	-0.0221	-0.0012
-2	-0.1111	.0217	-0.0038	-0.0101	.0029	-2	-0.1459	.0343	.0107	-0.0151	-0.0004
-1	-0.0667	.0139	-0.0024	-0.0047	.0036	-1	-0.0820	.0298	.0074	-0.0081	.0014
0	-0.0111	.0139	.0013	.0020	.0036	0	-0.0122	.0277	.0053	.0000	.0018
1	.0417	.0139	.0000	.0094	.0029	1	.0547	.0298	.0013	.0085	.0021
2	.0861	.0139	.0013	.0155	.0036	2	.1170	.0322	.0007	.0147	.0019
3	.1361	.0139	.0038	.0202	.0042	3	.1914	.0395	.0047	.0240	.0014
5	.2444	.0289	.0024	.0303	.0048	5	.3054	.0559	.0162	.0387	.0014
7	.3527	.0422	.0013	.0438	.0042	7	.4300	.082C	.0329	.0539	.0009
10	.5110	.0822	-0.0147	.0647	.0007	10	.6169	.1367	.0699	.0763	.0074
15	.7109	.2005	-0.1069	.0903	-0.0097	15	.9086	.2683	-0.1506	.1147	-0.0242
20	.7665	.3144	-0.1782	.0964	-0.0236	20	1.1456	.4409	-0.2178	.1457	-0.0484
25	.7637	.3905	-0.1806	.0944	-0.0333	25	1.2034	.5977	-0.2763	.1505	-0.0684
$M = 0.80$											
$M = 1.05$											
-10	-0.5343	.1016	-0.0125	-0.0616	-0.0127	-10	-0.6217	.1363	.0942	-0.0705	-0.0171
-7	-0.3875	.0500	-0.0116	-0.0438	-0.0059	-7	-0.4539	.0826	.0569	-0.0514	-0.0100
-5	-0.2765	.0297	-0.0099	-0.0297	-0.0026	-5	-0.3371	.0575	.0342	-0.0365	-0.0054
-3	-0.1750	.0184	-0.0092	-0.0402	-0.0005	-3	-0.2101	.0423	.0207	-0.0223	-0.0012
-2	-0.1204	.0158	-0.0066	-0.0110	.0005	-2	-0.1445	.0359	.0142	-0.0149	-0.0004
-1	-0.0677	.0120	-0.0033	-0.0046	.0011	-1	-0.0744	.0339	.0103	-0.0071	.0007
0	-0.0075	.0128	-0.0026	.0023	.0020	0	-0.0058	.0324	.0051	.0011	.0014
1	.0489	.0128	.0009	.0082	.0029	1	.0628	.0330	.0007	.0088	.0014
2	.0978	.0214	.0026	.0137	.0029	2	.1255	.0359	.0046	.0163	.0014
3	.1524	.0184	.0059	.0201	.0033	3	.1897	.0432	-0.0103	.0234	.0010
5	.2558	.0278	.0092	.0320	.0029	5	.3123	.0674	-0.0239	.0379	.0005
7	.3819	.0482	.0066	.0479	.0018	7	.4320	.0846	-0.0426	.0531	-0.0012
10	.5362	.0952	-0.0132	.0676	-0.0022	10	.6144	.1392	-0.0781	.0744	-0.0077
15	.7224	.2111	-0.0924	.0913	-0.0164	15	.9135	.2726	-0.1608	.1116	-0.0248
20	.7713	.3145	-0.1615	.0959	-0.0304	20	1.1587	.4477	-0.2299	.1438	-0.0496
25	.7976	.4116	-0.1806	.1009	-0.0438	25	1.2725	.6272	-0.2853	.1594	-0.0725
$M = 0.90$											
$M = 1.10$											
-10	-0.5448	.1050	.0259	-0.0631	-0.0138	-10	-0.6119	.1401	.0937	-0.0681	-0.0207
-7	-0.3999	.0550	.0008	-0.0445	-0.0062	-7	-0.4491	.0848	.0552	-0.0494	-0.0113
-5	-0.2966	.0320	-0.0096	-0.0320	-0.0037	-5	-0.3284	.0573	.0354	-0.0358	-0.0060
-3	-0.1816	.0190	-0.0088	-0.0182	-0.0004	-3	-0.2021	.0415	.0223	-0.0221	-0.0012
-2	-0.1233	.0163	-0.0073	-0.0121	.0006	-2	-0.1417	.0351	.0162	-0.0150	-0.0015
-1	-0.0616	.0147	-0.0029	-0.0049	.0023	-1	-0.0758	.0312	.0112	-0.0068	.0005
0	-0.0133	.0130	.0000	.0012	.0025	0	-0.0126	.0298	.0068	.0000	.0012
1	.0500	.0180	.0029	.0081	.0025	1	.0589	.0312	.0007	.0072	.0008
2	.1066	.0180	.0067	.0150	.0033	2	.1263	.0345	-0.0044	.0153	.0008
3	.1649	.0230	.0096	.0214	.0033	3	.1881	.0393	-0.0106	.0221	.0005
5	.2732	.0353	.0103	.0340	.0031	5	.3073	.0567	-0.0243	.0361	.0005
7	.3899	.0556	.0023	.0485	.0016	7	.4322	.0822	-0.0429	.0511	-0.0010
10	.5431	.1899	-0.0199	.0691	-0.0033	10	.6035	.1353	-0.0800	.0715	-0.0067
15	.7631	.2173	-0.0766	.0983	-0.0171	15	.8897	.2610	-0.1590	.1063	-0.0224
20	.8697	.3522	-0.1798	.1072	-0.0349	20	1.1367	.4348	-0.2291	.1373	-0.0455
25	.9230	.4702	-0.2094	.1132	-0.0510	25	1.2658	.6169	-0.2800	.1550	-0.0689

TABLE 2.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.20$$

$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$	$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$
$M = 0.40$											
$M = 0.55$											
-10	-1252					-10	-3692	.0965	-0.0474	-0.0407	-0.0095
-7	.0653					-7	-2254	.0559	-0.0721	-0.0233	-0.0039
-5	.0109					-5	-0.959	.0377	-0.0969	-0.0070	-0.0004
-3	.1034					-3	.0479	.0284	-0.1132	.0097	.0007
-2	.1469					-2	.1247	.0297	-0.1160	.0175	.0007
-1	.1905					-1	.1838	.0323	-0.1181	.0240	.0007
0	.2394					0	.2397	.0355	-0.1181	.0307	.0005
1	.2830					1	.3037	.0432	-0.1216	.0372	-0.0002
2	.3156					2	.3644	.0495	-0.1195	.0446	-0.0004
3	.3646					3	.4267	.0588	-0.1286	.0520	-0.0015
5	.4408					5	.5354	.0809	-0.1358	.0648	-0.0032
7	.5387					7	.6457	.1170	-0.1464	.0788	-0.0071
10	.6584					10	.7895	.1768	-0.1640	.0962	-0.0143
15	.7945					15	.9653	.2979	-0.1903	.1203	-0.0301
20	.8217					20	1.0708	.4481	-0.2552	.1280	-0.0487
25	.8108					25	1.0772	.5706	-0.2821	.1268	-0.0643
$M = 0.60$											
$M = 1.00$											
-10	-2016	.0633	-.0719	-.0204	-.0007	-10	-4180	.1093	-.0190	-.0468	-.0119
-7	-.0784	.0246	-.0954	-.0048	.0000	-7	-2603	.0723	-.0508	-.0260	-.0057
-5	.0196	.0123	-.0867	.0068	.0013	-5	-1347	.0527	-.0793	-.0111	-.0027
-3	.1176	.0140	-.0965	.0177	.0023	-3	-0.0046	.0438	-.0996	.0026	-.0005
-2	.1792	.0151	-.0891	.0245	.0039	-2	.0674	.0413	-.1077	.0115	-.0002
-1	.2240	.0179	-.0856	.0306	.0026	-1	.1501	.0429	-.1151	.0204	-.0002
0	.2632	.0246	-.1003	.0340	.0023	0	.2174	.0453	-.1219	.0279	-.0004
1	.3135	.0291	-.0990	.0414	.0020	1	.2787	.0511	-.1280	.0353	-.0005
2	.3611	.0342	-.0979	.0449	.0010	2	.3353	.0579	-.1320	.0409	-.0016
3	.4059	.0414	-.0954	.0476	.0010	3	.4042	.0677	-.1389	.0491	-.0029
5	.4871	.0565	-.0941	.0585	-.0003	5	.5206	.0903	-.1531	.0621	-.0039
7	.5991	.0868	-.0954	.0713	.0020	7	.6461	.1286	-.1721	.0766	-.0084
10	.7335	.1501	-.1028	.0877	-.0052	10	.8146	.1905	-.1930	.0985	-.0183
15	.8342	.2587	-.1635	.1012	-.0183	15	1.0626	.3375	-.2418	.1293	-.0360
20	.8566	.3651	-.1969	.1026	-.0332	20	1.2464	.5136	-.2790	.1524	-.0598
25	.8314	.4418	-.2280	.0979	-.0413	25	1.2617	.6673	-.3306	.1524	-.0785
$M = 0.80$											
$M = 1.05$											
-10	-2465	.0664	-.0856	-.0276	-.0049	-10	-4293	.1120	-.0085	-.0464	-.0115
-7	.1043	.0326	-.1065	-.0083	-.0018	-7	-2587	.0729	-.0481	-.0268	-.0063
-5	-.0057	.0243	-.1082	.0032	.0000	-5	.1411	.0541	-.0715	-.0125	-.0032
-3	.1062	.0607	-.1098	.0152	.0011	-3	-0.0888	.0456	-.0904	.0018	-.0012
-2	.1612	.0243	-.1098	.0216	.0018	-2	.0529	.0435	-.0989	.0104	-.0005
-1	.2276	.0281	-.1082	.0267	.0013	-1	.1235	.0447	-.1054	.0175	-.0010
0	.2788	.0300	-.1082	.0322	.0009	0	.1823	.0470	-.1118	.0239	-.0015
1	.3167	.0372	-.1057	.0391	.0013	1	.2499	.0529	-.1203	.0314	-.0021
2	.3679	.0448	-.1057	.0451	.0006	2	.3146	.0600	-.1236	.0382	-.0022
3	.4210	.0523	-.1048	.0506	-.0005	3	.3822	.0694	-.1327	.0457	-.0034
5	.5158	.0709	-.1007	.0612	.0009	5	.4998	.0926	-.1509	.0589	-.0048
7	.6296	.1009	-.1024	.0750	-.0031	7	.6292	.1273	-.1697	.0735	-.0091
10	.7567	.1642	-.1167	.0916	-.0093	10	.7880	.1902	-.2029	.0928	-.0174
15	.8382	.2666	-.1695	.1026	-.0229	15	1.0350	.3311	-.2504	.1245	-.0354
20	.8704	.3759	-.2115	.1036	-.0379	20	1.2261	.5178	-.2972	.1456	-.0603
25	.8761	.4646	-.2189	.1036	-.0498	25	1.3231	.7013	-.3330	.1609	-.0834
$M = 0.90$											
$M = 1.10$											
-10	-.3107	.0769	-.0661	-.0342	-.0057	-10	-4183	.1133	-.0062	-.0446	-.0128
-7	.1646	.0403	-.0966	-.0155	-.0024	-7	-2572	.0715	-.0419	-.0268	-.0079
-5	-.0470	.0272	-.1107	-.0021	.0006	-5	-1357	.0520	-.0712	-.0120	-.0044
-3	.0806	.0238	-.1166	.0122	.0014	-3	-.0113	.0438	-.0888	.0010	-.0020
-2	.1461	.0249	-.1166	.0187	.0014	-2	.0509	.0418	-.0963	.0082	-.0018
-1	.1999	.0289	-.1130	.0253	.0014	-1	.1159	.0430	-.1013	.0154	-.0013
0	.2570	.0329	-.1137	.0326	.0008	0	.1724	.0367	-.1088	.0240	-.0018
1	.3141	.0396	-.1166	.0391	.0006	1	.2459	.0458	-.1157	.0299	-.0028
2	.3678	.0464	-.1166	.0452	-.0002	2	.3166	.0557	-.1231	.0357	-.0033
3	.4316	.0561	-.1130	.0522	-.0006	3	.3731	.0667	-.1313	.0429	-.0044
5	.5341	.0769	-.1166	.0636	-.0022	5	.4918	.0890	-.1456	.0563	-.0056
7	.6392	.1115	-.1241	.0775	-.0053	7	.6190	.1244	-.1682	.0710	-.0102
10	.7541	.1676	-.1389	.0925	-.0121	10	.7830	.1843	-.1945	.0892	-.0182
15	.9304	.2832	-.1612	.1199	-.0283	15	1.0232	.3219	-.2451	.1184	-.0363
20	.9741	.4155	-.2393	.1158	-.0443	20	1.2154	.5045	-.2951	.1423	-.0595
25	.9674	.5253	-.2526	.1158	-.0580	25	1.3257	.6894	-.3326	.1564	-.0861

TABLE 2.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.1089					-10	-0.2939	.0936	-0.0834	-0.0349	-0.0076
-7	-0.0109					-7	-0.1342	.0549	-0.1131	-0.0140	-0.0024
-5	.0708					-5	-0.0096	.0425	-0.1301	0.0000	0.0002
-3	.1634					-3	.1326	.0383	-0.1427	.0155	.0015
-2	.2015					-2	.2012	.0425	-0.1497	.0233	.0009
-1	.2559					-1	.2699	.0441	-0.1441	.0314	.0005
0	.2995					0	.3306	.0502	-0.1469	.0384	.0000
1	.3321					1	.3929	.0588	-0.1541	.0450	-0.0005
2	.3866					2	.4536	.0674	-0.1541	.0523	-0.0013
3	.4247					3	.5175	.0776	-0.1526	.0593	-0.0022
5	.5009					5	.6213	.1038	-0.1541	.0717	-0.0048
7	.6044					7	.7267	.1437	-0.1681	.0853	-0.0096
10	.7405					10	.8752	.2035	-0.1823	.1027	-0.0180
15	.8494					15	1.0541	.3348	-0.2035	.1279	-0.0357
20	.8715					20	1.1627	.4932	-0.2636	.1384	-0.0561
25	.8385					25	1.1468	.6191	-0.2918	.1318	-0.0719
$M = 0.60$											
$M = 1.00$											
-10	-0.1540	.0661	-.0793	-.0204	-.0026	-10	-0.3549	.1046	-.0582	-.0408	-.0098
-7	-0.0224	.0302	-.1015	-.0027	-.0010	-7	-0.1897	.0691	-.0880	-.0212	-.0041
-5	.0812	.0218	-.0966	.0102	-.0003	-5	-0.0642	.0557	-.1137	-.0074	-.0013
-3	.1708	.0190	-.1028	.0204	-.0007	-3	.0918	.0480	-.1340	.0111	0.0000
-2	.2240	.0207	-.1004	.0265	-.0007	-2	.1683	.0480	-.1442	.0197	.0004
-1	.2744	.0235	-.0916	.0319	-.0003	-1	.2432	.0541	-.1516	.0271	.0004
0	.3332	.0274	-.0930	.0374	-.0003	0	.3090	.0578	-.1591	.0345	-.0004
1	.3724	.0342	-.1028	.0428	-.0003	1	.3793	.0661	-.1624	.0423	-.0005
2	.4173	.0370	-.0916	.0476	-.0007	2	.4436	.0743	-.1651	.0497	-.0014
3	.4649	.0498	-.0986	.0537	-.0007	3	.5124	.0872	-.1746	.0575	-.0023
5	.5489	.0672	-.0966	.0632	-.0010	5	.6134	.1120	-.1848	.0705	-.0048
7	.6581	.0952	-.0842	.0748	-.0039	7	.7403	.1557	-.1996	.0854	-.0101
10	.7925	.1557	-.1004	.0917	-.0108	10	.8994	.2233	-.2220	.1040	-.0190
15	.8709	.2711	-.1709	.1019	-.0248	15	1.1380	.3723	-.2626	.1337	-.0397
20	.8989	.3842	-.2229	.1054	-.0381	20	1.3124	.5552	-.2964	.1578	-.0658
25	.8625	.4598	-.2267	.0972	-.0476	25	1.3277	.7058	-.3357	.1578	-.1039
$M = 0.80$											
$M = 1.05$											
-10	-0.1821	.0577	-.1007	-.0230	-.0024	-10	-0.3776	.1070	-.0442	-.0410	-.0099
-7	-0.0436	.0288	-.1141	-.0046	-.0005	-7	-0.1925	.0702	-.0799	-.0217	-.0051
-5	.0645	.0243	-.1107	.0078	.0011	-5	-0.0749	.0570	-.1059	-.0089	-.0019
-3	.1707	.0250	-.1174	.0203	.0016	-3	.0690	.0491	-.1254	.0075	-.0005
-2	.2295	.0269	-.1158	.0272	.0006	-2	.1454	.0500	-.1339	.0160	.0004
-1	.2864	.0300	-.1074	.0336	.0006	-1	.1988	.0447	-.1417	.0232	.0002
0	.3357	.0364	-.1115	.0387	.0000	0	.2733	.0549	-.1403	.0125	-.0004
1	.3982	.0440	-.1200	.0460	-.0005	1	.3438	.0643	-.1547	.0374	-.0021
2	.4438	.0504	-.1167	.0511	-.0006	2	.4084	.0729	-.1611	.0453	-.0021
3	.4874	.0588	-.1141	.0562	-.0016	3	.4745	.0861	-.1716	.0531	-.0031
5	.5860	.0793	-.1115	.0677	-.0029	5	.5891	.1134	-.1859	.0663	-.0063
7	.7055	.1130	-.1098	.0819	-.0071	7	.7067	.1516	-.2002	.0810	-.0106
10	.8287	.1726	-.1200	.0985	-.0146	10	.8785	.2210	-.2230	.1009	-.0197
15	.9254	.2852	-.1602	.1105	-.0293	15	1.1077	.3743	-.2639	.1302	-.0400
20	.9368	.4020	-.2206	.1086	-.0437	20	1.2870	.5577	-.3106	.1533	-.0642
25	.9254	.4980	-.2241	.1082	-.0558	25	1.3927	.7631	-.3509	.1676	-.0912
$M = 0.90$											
$M = 1.10$											
-10	-0.2493	.0773	-.1007	-.0305	-.0041	-10	-0.3589	.1085	-.0413	-.0405	-.0120
-7	-0.0920	.0418	-.1206	-.0106	-.0010	-7	-0.1808	.0687	-.0818	-.0209	-.0064
-5	.0234	.0298	-.1281	.0041	.0014	-5	-0.0650	.0543	-.1031	-.0075	-.0036
-3	.1474	.0298	-.1288	.0175	.0016	-3	.0692	.0480	-.1206	-.0075	-.0012
-2	.2109	.0338	-.1281	.0240	.0014	-2	.1356	.0480	-.1300	.0148	-.0020
-1	.2661	.0378	-.1296	.0305	.0006	-1	.2006	.0494	-.1312	.0223	-.0026
0	.3247	.0445	-.1340	.0370	.0000	0	.2600	.0548	-.1456	.0288	-.0033
1	.3835	.0519	-.1317	.0443	-.0004	1	.3292	.0633	-.1531	.0363	-.0033
2	.4505	.0609	-.1347	.0516	-.0014	2	.3843	.0709	-.1569	.0429	-.0051
3	.5087	.0733	-.1355	.0589	-.0019	3	.4479	.0814	-.1643	.0504	-.0061
5	.6091	.0954	-.1317	.0707	-.0043	5	.5651	.1085	-.1788	.0634	-.0077
7	.7146	.1332	-.1428	.0841	-.0084	7	.6909	.1486	-.1894	.0778	-.0137
10	.8434	.1901	-.1466	.0999	-.0160	10	.8449	.2139	-.2175	.0957	-.0222
15	1.0041	.3119	-.1762	.1219	-.0323	15	1.0709	.3628	-.2581	.1238	-.0414
20	1.0242	.4428	-.2443	.1198	-.0487	20	1.2546	.5462	-.3038	.1464	-.0660
25	1.0242	.5529	-.2613	.1190	-.0636	25	1.3676	.7533	-.3476	.1612	-.0936

TABLE 2.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 2 MODEL - Concluded

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.55$											
-10	-1077					-10	2504	.0785	-0.0974	-0.0290	-0.0052
-7	.0000					-7	0829	.0456	-0.1256	-0.0085	-0.0026
-5	.0916					-5	0447	.0345	-0.1411	.0070	-0.0032
-3	.1886					-3	1643	.0345	-0.1447	.0197	-0.0034
-2	.2370					-2	2265	.0376	-0.1481	.0271	-0.0037
-1	.2694					-1	2823	.0440	-0.1503	.0333	-0.0043
0	.3235					0	3429	.0501	-0.1481	.0399	-0.0050
1	.3609					1	4099	.0581	-0.1517	.0472	-0.0057
2	.4040					2	4626	.0689	-0.1539	.0538	-0.0067
3	.4418					3	5343	.0817	-0.1573	.0616	-0.0080
5	.5387					5	6300	.1097	-0.1659	.0743	-0.0113
7	.6411					7	7369	.1474	-0.1743	.0879	-0.0152
10	.7650					10	8773	.2166	-0.1905	.1045	-0.0232
15	.8515					15	1.0591	.3452	-0.2047	.1289	-0.0397
20	.8727					20	1.1261	.4941	-0.2660	.1347	-0.0579
25	.8189					25	1.1229	.6214	-0.2964	.1312	-0.0734
$M = 0.60$											
$M = 1.00$											
-10	-1448	.0479	-.0838	-.0176	-.0039	-10	3102	.0947	-.0730	-.0360	-.0076
-7	.0139	.0206	-.0876	-.0027	-.0039	-7	1329	.0617	-.1109	-.0148	-.0041
-5	.0919	.0011	-.0900	.0095	-.0036	-5	0000	.0480	-.1339	.0011	-.0048
-3	.1866	.0167	-.0911	.0196	-.0032	-3	1406	.0452	-.1521	.0178	-.0050
-2	.2423	.0189	-.0925	.0284	-.0023	-2	2185	.0452	-.1575	.0260	-.0059
-1	.2785	.0189	-.0862	.0338	-.0023	-1	2781	.0526	-.1649	.0323	-.0068
0	.3342	.0273	-.0925	.0379	-.0029	0	3515	.0571	-.1656	.0397	-.0076
1	.3844	.0340	-.0925	.0426	-.0032	1	4233	.0660	-.1676	.0471	-.0083
2	.4317	.0412	-.0887	.0500	-.0036	2	4691	.0752	-.1691	.0542	-.0098
3	.4735	.0479	-.0900	.0534	-.0045	3	5348	.0902	-.1744	.0620	-.0114
5	.5905	.0668	-.0925	.0656	-.0065	5	6448	.1201	-.1825	.0742	-.0151
7	.6824	.1003	-.0925	.0784	-.0094	7	7488	.1608	-.1961	.0875	-.0197
10	.8025	.1616	-.1110	.0940	-.0104	10	.8924	.2329	-.2197	.1057	-.0291
15	.8468	.2808	-.1861	.1001	-.0233	15	1.1094	.3802	-.2568	.1339	-.0316
20	.8746	.3972	-.2280	.1014	-.0376	20	1.2652	.5486	-.2955	.1539	-.0725
25	.8301	.4657	-.2317	.0919	-.0457	25	1.2469	.6913	-.3380	.1498	-.0888
$M = 0.80$											
$M = 1.05$											
-10	-1717	.0483	-.1002	-.0197	-.0033	-10	3155	.0925	-.0701	-.0360	-.0132
-7	.0245	.0249	-.1109	-.0019	-.0026	-7	1468	.0608	-.1026	-.0160	-.0087
-5	.0830	.0215	-.1127	.0096	-.0024	-5	0220	.0505	-.1266	-.0021	-.0065
-3	.1886	.0249	-.1127	.0220	-.0024	-3	1248	.0476	-.1448	.0142	-.0056
-2	.2414	.0298	-.1127	.0302	-.0026	-2	1981	.0505	-.1532	.0221	-.0060
-1	.3018	.0351	-.1102	.0357	-.0033	-1	2495	.1993	-.1571	.0278	-.0072
0	.3508	.0419	-.1127	.0417	-.0035	0	3185	.0619	-.1624	.0353	-.0077
1	.4055	.0483	-.1109	.0476	-.0044	1	3875	.0722	-.1694	.0435	-.0092
2	.4489	.0573	-.1102	.0540	-.0057	2	4506	.0793	-.1720	.0502	-.0101
3	.5036	.0668	-.1059	.0586	-.0059	3	5019	.0910	-.1766	.0570	-.0118
5	.6017	.0909	-.1043	.0705	-.0072	5	6238	.1227	-.1870	.0712	-.0152
7	.7168	.1279	-.1068	.0851	-.0103	7	.7456	.1632	-.2012	.0848	-.0203
10	.8394	.1894	-.1160	.0998	-.0178	10	.9011	.2339	-.2259	.1019	-.0297
15	.8696	.2950	-.1820	.1057	-.0303	15	1.1242	.3883	-.2695	.1322	-.0505
20	.9092	.4082	-.2187	.1062	-.0456	20	1.2651	.5630	-.3052	.1528	-.0737
25	.8884	.4991	-.2270	.1044	-.0572	25	1.3649	.3666	-.3506	.1642	-.0986
$M = 0.90$											
$M = 1.10$											
-10	-2204	.0738	-.1049	-.0239	-.0045	-10	3105	.0957	-.0656	-.0353	-.0143
-7	-.0584	.0417	-.1256	-.0057	-.0037	-7	1440	.0675	-.1012	-.0168	-.0095
-5	.0635	.0344	-.1329	.0085	-.0033	-5	0183	.0556	-.1249	-.0027	-.0074
-3	.1753	.0354	-.1329	.0215	-.0033	-3	1059	.0556	-.1387	.0117	-.0074
-2	.2371	.0401	-.1323	.0288	-.0041	-2	1821	.0590	-.1468	.0192	-.0075
-1	.2906	.0451	-.1315	.0361	-.0050	-1	2399	.0624	-.1498	.0260	-.0085
0	.3473	.0534	-.1293	.0422	-.0058	0	3063	.0680	-.1580	.0339	-.0093
1	.4091	.0615	-.1323	.0478	-.0060	1	3613	.0762	-.1617	.0408	-.0103
2	.4592	.0691	-.1293	.0551	-.0074	2	4319	.0867	-.1680	.0480	-.0115
3	.5160	.0805	-.1270	.0624	-.0089	3	4898	.0985	-.1735	.0552	-.0131
5	.6265	.1052	-.1293	.0746	-.0101	5	6041	.1270	-.1842	.0682	-.0164
7	.7214	.1429	-.1352	.0875	-.0136	7	.7198	.1694	-.1999	.0819	-.0217
10	.8617	.2054	-.1478	.1046	-.0217	10	.8609	.2340	-.2186	.0980	-.0299
15	1.0153	.3327	-.1832	.1265	-.0388	15	1.0727	.3706	-.2591	.1254	-.0489
20	1.0020	.4502	-.2453	.1200	-.0532	20	1.2279	.5428	-.3122	.1459	-.0724
25	.9886	.5568	-.2586	.1180	-.0664	25	1.3267	.7218	-.3435	.1579	-.0955

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL

$$n_{\text{cl}} = 0.04$$

$\frac{C_f}{C} = \text{NONE}$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.5838					-10	-0.7947	+0.1667	+0.0846	-0.0958	+0.197
-7	-0.4338					-7	-0.6088	+0.0971	+0.0540	-0.0726	+0.114
-5	-0.3191					-5	-0.4410	+0.0546	+0.0200	-0.0509	+0.065
-3	-0.2104					-3	-0.2879	+0.0304	+0.0049	-0.0326	+0.035
-2	-0.1544					-2	-0.2057	+0.0211	+0.0006	-0.0226	+0.026
-1	-0.0956					-1	-0.1218	+0.0158	-0.0036	-0.0133	+0.021
0	-0.0279					0	-0.0296	+0.0129	+0.0038	-0.0030	+0.019
1	0.0176					1	-0.0609	+0.0129	+0.0097	+0.0073	+0.025
2	0.0691					2	+0.1431	+0.0162	+0.0119	+0.0163	+0.030
3	0.1250					3	+0.2287	+0.0239	+0.0108	+0.0270	+0.043
5	0.2323					5	+0.4048	+0.0453	-0.0085	+0.0479	+0.079
7	0.3426					7	+0.5709	+0.0810	-0.0418	+0.0689	+0.132
10	0.5382					10	+0.7848	+1.505	-0.0769	+0.095	+0.238
15	0.6661					15	1.0530	+0.2930	-0.1140	+0.1305	+0.458
20	0.6941										
25	0.6764										
$M = 0.60$											
-10	-0.6694	+0.1170	+0.0142	-0.0804	+0.0156	-10	-0.7981	+0.1663	+0.1063	-0.0918	+0.190
-7	-0.5137	+0.0601	-0.0208	-0.0580	+0.0082	-7	-0.5884	+0.0962	+0.0623	-0.0658	+0.019
-5	-0.3711	+0.0322	-0.0196	-0.0408	+0.0047	-5	-0.4382	+0.0604	+0.0377	-0.0491	+0.071
-3	-0.2474	+0.0204	-0.0084	-0.0263	+0.0031	-3	-0.2911	+0.0393	+0.0234	-0.0317	+0.043
-2	-0.1746	+0.0162	-0.0066	-0.0183	+0.0024	-2	-0.2034	+0.0297	+0.0122	-0.0218	+0.033
-1	-0.1019	+0.0140	-0.0006	-0.0115	+0.0018	-1	-0.1189	+0.0231	+0.0025	-0.0120	+0.027
0	-0.0335	+0.0122	+0.0035	-0.0028	+0.0017	0	-0.0219	+0.0200	-0.0019	-0.0022	+0.022
1	0.0291	+0.0122	+0.0007	+0.0053	+0.0019	1	+0.0782	+0.0208	+0.0058	+0.0085	+0.029
2	0.0975	+0.0162	+0.0078	+0.0128	+0.0026	2	+0.1721	+0.0239	-0.0095	+0.0193	+0.039
3	0.1673	+0.0218	+0.0123	+0.0208	+0.0037	3	+0.2504	+0.0315	+0.0170	+0.0291	+0.050
5	0.2983	+0.0373	+0.0235	+0.0364	+0.0055	5	+0.4163	+0.0523	-0.0368	+0.0481	+0.084
7	0.4264	+0.0633	+0.0223	+0.0518	+0.0096	7	+0.5571	+0.0816	+0.0555	+0.0639	+0.127
10	0.6039	+1.224	-0.0071	+0.0762	+0.0181	10	+0.7668	+1.451	-0.0956	+0.0889	+0.222
15	0.7334	+2.269	-0.0691	+0.0963	+0.0327	15	1.0485	+2.901	-0.1314	+0.1272	+0.452
20	0.7421	+3.070	-0.0860	+0.0971	+0.0441						
25	0.7480	+3.922	-0.0869	+0.0971	+0.0539						
$M = 0.80$											
-10	-0.7036	+0.1297	+0.0039	-0.0835	+0.0162	-10	-0.7557	+0.1550	+0.0966	-0.0865	+0.174
-7	-0.5534	+0.0695	-0.0125	-0.0626	+0.0186	-7	-0.5592	+0.0911	+0.0601	-0.0630	+0.100
-5	-0.4131	+0.0384	-0.0165	-0.0456	+0.0055	-5	-0.4232	+0.0583	+0.0390	-0.0468	+0.060
-3	-0.2727	+0.0223	-0.0098	-0.0294	+0.0034	-3	-0.2811	+0.0376	+0.0226	-0.0303	+0.035
-2	-0.1878	+0.0160	-0.0066	-0.0200	+0.0026	-2	-0.1965	+0.0286	+0.0125	-0.0208	+0.025
-1	-0.1087	+0.0131	-0.0028	-0.0113	+0.0020	-1	-0.1058	+0.0230	+0.0023	+0.0110	+0.018
0	-0.0395	+0.0107	+0.0033	-0.0038	+0.0018	0	-0.0212	+0.0212	-0.0018	-0.0040	+0.016
1	0.0296	+0.0102	+0.0063	+0.0047	+0.0021	1	+0.0725	+0.0215	+0.0057	+0.0089	+0.023
2	0.1127	+0.0141	+0.0127	+0.0142	+0.0030	2	+0.1602	+0.0260	+0.0172	+0.0180	+0.032
3	0.1917	+0.0190	+0.0137	+0.0227	+0.0036	3	+0.2418	+0.0335	+0.0190	+0.0275	+0.043
5	0.3498	+0.0340	+0.0269	+0.0407	+0.0067	5	+0.4020	+0.0535	-0.0386	+0.0456	+0.072
7	0.4961	+0.0613	+0.0226	+0.0584	+0.0111	7	+0.5350	+0.0825	-0.0554	+0.0618	+0.118
10	0.6818	+1.186	+0.0068	+0.0823	+0.0192	10	+0.7254	+1.420	-0.0912	+0.0853	+0.210
15	0.7609	+2.221	-0.0697	+0.0987	+0.0341						
20	0.7767	+3.091	-0.0925	+0.0991	+0.0438						
25	0.7905	+4.004	-0.0985	+0.1007	+0.0576						
$M = 0.90$											
-10	-0.7338	+0.1444	+0.0318	-0.0903	+0.0177	-10	-0.7320	+0.1512	+0.0978	-0.0838	+0.162
-7	-0.5755	+0.0823	+0.0112	-0.0680	+0.0099	-7	-0.5417	+0.0897	+0.0561	-0.0616	+0.093
-5	-0.4358	+0.0458	-0.0037	-0.0494	+0.0057	-5	-0.4246	+0.0583	+0.0423	-0.0456	+0.056
-3	-0.2963	+0.0244	-0.0101	-0.0324	+0.0033	-3	-0.2782	+0.0385	+0.0254	-0.0296	+0.034
-2	-0.2004	+0.0159	-0.0106	-0.0215	+0.0026	-2	-0.1903	+0.0295	+0.0146	-0.0201	+0.026
-1	-0.1220	+0.0116	-0.0056	-0.0130	+0.0020	-1	-0.1083	+0.0242	+0.0069	-0.0107	+0.019
0	-0.0366	+0.0094	+0.0019	-0.0039	+0.0018	0	-0.0234	+0.0220	+0.0010	+0.0278	+0.017
1	0.0488	+0.0085	+0.0095	+0.0064	+0.0021	1	+0.0673	+0.0216	-0.0080	+0.0083	+0.023
2	0.1290	+0.0116	+0.0159	+0.0148	+0.0028	2	+0.1464	+0.0266	+0.0125	+0.0169	+0.029
3	0.2266	+0.0172	+0.0179	+0.0254	+0.0038	3	+0.2284	+0.0342	+0.0194	+0.0267	+0.038
5	0.4009	+0.0386	+0.0078	+0.0469	+0.0071	5	+0.3806	+0.0558	+0.0373	+0.0438	+0.063
7	0.5438	+0.0707	-0.0094	+0.0659	+0.0117	7	+0.5124	+0.0853	-0.0548	+0.0598	+0.101
10	0.7146	+1.294	-0.0273	+0.0896	+0.0206	10	+0.6881	+1.465	-0.0868	+0.0832	+0.176
15	0.8245	+2.418	-0.0846	+1.054	+0.0354						
20	0.8715	+3.403	-0.1196	+1.082	+0.0481						
25	0.8820	+4.406	-0.1259	+1.089	+0.0590						
$M = 1.00$											
-10	-0.7947	+0.1667	+0.0846	-0.0958	+0.0177	-10	-0.7320	+0.1512	+0.0978	-0.0838	+0.162
-7	-0.6088	+0.0971	+0.0540	-0.0726	+0.0114	-7	-0.5417	+0.0897	+0.0561	-0.0616	+0.093
-5	-0.4410	+0.0546	+0.0200	-0.0509	+0.0065	-5	-0.4246	+0.0583	+0.0423	-0.0456	+0.056
-3	-0.2879	+0.0304	+0.0049	-0.0326	+0.0035	-3	-0.2782	+0.0385	+0.0254	-0.0296	+0.034
-2	-0.2057	+0.0211	+0.0006	-0.0226	+0.0026	-2	-0.1903	+0.0295	+0.0146	-0.0201	+0.026
-1	-0.1218	+0.0158	-0.0036	-0.0133	+0.0021	-1	-0.1083	+0.0242	+0.0069	-0.0107	+0.019
0	-0.0296	+0.0129	+0.0038	-0.0030	+0.0019	0	-0.0234	+0.0220	+0.0010	+0.0278	+0.017
1	0.0609	+0.0129	+0.0097	+0.0073	+0.0021	1	+0.0673	+0.0216	-0.0080	+0.0083	+0.023
2	0.1431	+0.0162	+0.0119	+0.0163	+0.0030	2	+0.1464	+0.0266	+0.0125	+0.0169	+0.029
3	0.2287	+0.0239	+0.0108	+0.0270	+0.0043	3	+0.2284	+0.0342	+0.0194	+0.0267	+0.038
5	0.4048	+0.0453	-0.0085	+0.0479	+0.0079	5	+0.3806	+0.0558	+0.0373	+0.0438	+0.063
7	0.5709	+0.0810	-0.0418	+0.0689	+0.0132	7	+0.5124	+0.0853	-0.0548	+0.0598	+0.101
10	0.7848	+1.505	-0.0769	+0.095	+0.0238	10	+0.6881	+1.465	-0.0868	+0.0832	+0.176
15	1.0530	+2.930	-0.1140	+1.305	+0.0458						

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{c_f}{c} = 0.10$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
M = 0.40											
M = 0.55											
-10	-0.3945					-10	-0.6736	0.1520	0.0242	-0.0797	0.0193
-7	-0.2221					-7	-0.4643	0.0864	-0.0214	-0.0532	0.0113
-5	-0.1125					-5	-0.2894	0.0515	-0.0593	-0.0317	0.0071
-3	-0.0132					-3	-0.1341	0.0342	-0.0850	-0.0127	0.0051
-2	0.0409					-2	-0.0523	0.0285	-0.0934	-0.0020	0.0048
-1	0.0906					-1	0.0589	0.0253	-0.1109	0.0101	0.0044
0	0.1476					0	0.1602	0.0314	-0.1137	0.0212	0.0052
1	0.1973					1	0.2420	0.0369	-0.1185	0.0306	0.0063
2	0.2440					2	0.3564	0.0406	-0.1179	0.0435	0.0075
3	0.2878					3	0.4381	0.0522	-0.1250	0.0534	0.0094
5	0.3945					5	0.5984	0.0836	-0.1377	0.0716	0.0138
7	0.5085					7	0.7291	0.1206	-0.1475	0.0886	0.0191
10	0.6765										
15	0.7978										
20	0.7978										
25	0.7598										
M = 0.60											
M = 1.00											
-10	-0.4583	0.0828	-0.0647	-0.0544	0.0126	-10	-0.6488	0.1484	0.0366	-0.0134	0.0179
-7	-0.2776	0.0437	-0.0958	-0.0310	0.0073	-7	-0.4621	0.0899	-0.0078	-0.0515	0.0114
-5	-0.1403	0.0288	-0.0976	-0.0135	0.0048	-5	-0.3112	0.0604	-0.0365	-0.0334	0.0082
-3	-0.0174	0.0253	-0.0845	0.0004	0.0040	-3	-0.1774	0.0443	-0.0532	-0.0176	0.0063
-2	0.0549	0.0253	-0.0867	0.0089	0.0040	-2	-0.0918	0.0379	-0.0654	-0.0074	0.0057
-1	0.1128	0.0278	-0.0832	0.0158	0.0042	-1	-0.0109	0.0356	-0.0694	0.0020	0.0052
0	0.1807	0.0348	-0.0764	0.0234	0.0048	0	0.0794	0.0363	-0.0792	0.0115	0.0054
1	0.2400	0.0431	-0.0742	0.0314	0.0059	1	0.1774	0.0387	-0.0888	0.0233	0.0063
2	0.3181	0.0483	-0.0719	0.0399	0.0074	2	0.2738	0.0443	-0.0981	0.0338	0.0072
3	0.3803	0.0562	-0.0668	0.0472	0.0087	3	0.3672	0.0528	-0.1109	0.0449	0.0087
5	0.5176	0.0814	-0.0527	0.0633	0.0114	5	0.5493	0.0815	-0.1356	0.0658	0.0125
7	0.6405	0.1160	-0.0497	0.0798	0.0163	7	0.6893	0.1167	-0.1518	0.0828	0.0173
10	0.7895	0.1835	-0.0756	0.1012	0.0255						
15	0.8502	0.2880	-0.1271	0.1129	0.0391						
20	0.8473	0.3741	-0.1383	0.1106	0.0496						
25	0.8328	0.4594	-0.1413	0.1100	0.0601						
M = 0.80											
M = 1.05											
-10	-0.5045	0.0939	-0.0769	-0.0619	0.0128	-10	-0.6371	0.1434	0.0351	-0.0760	0.0156
-7	-0.3278	0.0488	-0.0946	-0.0381	0.0071	-7	-0.4478	0.0890	-0.0071	-0.0508	0.0089
-5	-0.1845	0.0273	-0.1016	-0.0203	0.0049	-5	-0.3186	0.0609	-0.0320	-0.0347	0.0058
-3	-0.0394	0.0200	-0.0947	0.0038	0.0038	-3	-0.1863	0.0462	-0.0456	-0.0182	0.0040
-2	0.0354	0.0188	-0.0944	0.0056	0.0036	-2	-0.1037	0.0403	-0.0580	-0.0082	0.0040
-1	0.1139	0.0205	-0.0962	0.0145	0.0040	-1	-0.0120	0.0373	-0.0645	0.0009	0.0027
0	0.2022	0.0278	-0.0864	0.0244	0.0047	0	0.0796	0.0388	-0.0731	0.0116	0.0036
1	0.2670	0.0296	-0.0912	0.0332	0.0056	1	0.1548	0.0406	-0.0808	0.0207	0.0040
2	0.3494	0.0382	-0.0849	0.0427	0.0068	2	0.2449	0.0466	-0.0878	0.0307	0.0053
3	0.4378	0.0478	-0.0789	0.0522	0.0080	3	0.3351	0.0565	-0.0981	0.0410	0.0068
5	0.5791	0.0715	-0.0715	0.0695	0.0121	5	0.5049	0.0842	-0.1246	0.0617	0.0105
7	0.7185	0.1062	-0.0718	0.0870	0.0174	7	0.6446	0.1186	-0.1473	0.0805	0.0149
10	0.8598	0.1704	-0.0769	0.1096	0.0261						
15	0.8520	0.2733	-0.1409	0.1112	0.0375						
M = 0.90											
M = 1.10											
-10	-0.5787	0.1206	-0.0358	-0.0708	0.0156	-10	-0.6145	0.1389	0.0979	-0.0736	0.0136
-7	-0.3846	0.0651	-0.0707	-0.0449	0.0086	-7	-0.4470	0.0870	0.0542	-0.0507	0.0075
-5	-0.2426	0.0375	-0.0863	-0.0259	0.0057	-5	-0.3174	0.0594	0.0244	-0.0336	0.0044
-3	-0.0866	0.0243	-0.0993	-0.0082	0.0043	-3	-0.1849	0.0459	0.0019	-0.0183	0.0027
-2	0.0055	0.0204	-0.1031	0.0028	0.0039	-2	-0.0946	0.0412	-0.0132	-0.0085	0.0020
-1	0.0918	0.0204	-0.1057	0.0128	0.0042	-1	-0.0291	0.0369	-0.0264	0.0000	0.0020
0	0.1836	0.0260	-0.1001	0.0233	0.0048	0	0.0612	0.0394	-0.0415	0.0097	0.0021
1	0.2824	0.0302	-0.1066	0.0347	0.0059	1	0.1427	0.0419	-0.0570	0.0200	0.0039
2	0.3794	0.0388	-0.1013	0.0456	0.0074	2	0.2359	0.0465	-0.0708	0.0292	0.0042
3	0.4608	0.0464	-0.1027	0.0548	0.0090	3	0.3203	0.0566	-0.0881	0.0395	0.0054
5	0.6205	0.0771	-0.1087	0.0741	0.0135	5	0.4732	0.0852	-0.1184	0.0583	0.0082
7	0.7467	0.1163	-0.1232	0.0922	0.0191	7	0.6116	0.1193	-0.1463	0.0751	0.0120
10	0.9390	0.1866	-0.1353	0.1156	0.0291						

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Continued

$$\frac{k}{c} = 0.04 \quad \frac{C_f}{c} = 0.20$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.3057					-10	-0.5765	0.1362	-0.0406	-0.0702	0.0144
-7	-0.1254					-7	-0.3423	0.0770	0.2230	-0.0399	0.0069
-5	-0.0029					-5	-0.1809	0.0501	-0.1155	-0.0180	0.0038
-3	0.1081					-3	0.0129	0.0362	-0.1487	0.0049	0.0028
-2	0.1629					-2	0.1179	0.0342	-0.1631	0.0163	0.0030
-1	0.2206					-1	0.2228	0.0354	-0.1689	0.0284	0.0038
0	0.2783					0	0.3246	0.0409	-0.1713	0.0399	0.0048
1	0.3258					1	0.4231	0.0480	-0.1716	0.0506	0.0063
2	0.4023					2	0.5167	0.0580	-0.1707	0.0611	0.0083
3	0.4369					3	0.6071	0.0706	-0.1712	0.0725	0.0106
5	0.5911					5	0.7638	0.1056	-0.1845	0.0905	0.0159
7	0.6762										
10	0.8074										
15	0.8708										
20	0.8449										
25	0.8160										
$M = 0.60$											
-10	-0.3554	0.0688	-0.0808	-0.0426	0.0056	-10	-0.5945	0.1420	0.2813	-0.0699	0.0146
-7	-0.1513	0.0267	-0.1203	-0.0165	0.0009	-7	-0.3871	0.0873	-0.0571	-0.0435	0.0079
-5	0.0014	0.0137	-0.1212	0.0014	-0.0002	-5	-0.2289	0.0608	-0.0848	-0.0239	0.0050
-3	0.1327	0.0154	-0.1174	0.0169	0.0001	-3	-0.0615	0.0446	-0.1145	-0.0037	0.0037
-2	0.2027	0.0197	-0.1144	0.0258	0.0015	-2	0.0307	0.0408	-0.1271	0.0078	0.0034
-1	0.2683	0.0243	-0.1072	0.0346	0.0043	-1	0.1352	0.0400	-0.1419	0.0180	0.0034
0	0.3325	0.0320	-0.0977	0.0433	0.0067	0	0.2842	0.0400	-0.1645	0.0357	0.0045
1	0.3982	0.0414	-0.0946	0.0504	0.0089	1	0.3656	0.0465	-0.1668	0.0451	0.0058
2	0.4538	0.0477	-0.0830	0.0577	0.0112	2	0.4701	0.0571	-0.1698	0.0566	0.0077
3	0.5233	0.0561	-0.0830	0.0665	0.0137	3	0.5531	0.0691	-0.1747	0.0665	0.0098
5	0.6494	0.0846	-0.0577	0.0837	0.0195	5	0.7082	0.1024	-0.1858	0.0848	0.0147
7	0.7792	0.1200	-0.0609	0.0996	0.0263						
10	0.8934	0.1845	-0.0965	0.1155	0.0335						
15	0.9220	0.2850	-0.1443	0.1175	0.0405						
20	0.9077	0.3706	-0.1471	0.1140	0.0498						
25	0.8906	0.4577	-0.1484	0.1126	0.0607						
$M = 0.80$											
-10	-0.3741	0.0817	-0.0904	-0.0902	0.0019	-10	-0.5814	0.1360	-0.0138	-0.0684	0.0134
-7	-0.1861	0.0372	-0.1360	-0.0392	0.0026	-7	-0.3767	0.0839	-0.0536	-0.0420	0.0073
-5	-0.0233	0.0212	-0.1379	-0.0010	0.0009	-5	-0.2254	0.0598	-0.0808	-0.0240	0.0044
-3	0.1153	0.0188	-0.1342	0.0314	0.0008	-3	-0.0727	0.0460	-0.1039	-0.0054	0.0031
-2	0.2045	0.0198	-0.1350	0.0506	0.0015	-2	0.0193	0.0423	-0.1160	0.0051	0.0029
-1	0.2791	0.0234	-0.1297	0.0676	0.0030	-1	0.1009	0.0412	-0.1275	0.0150	0.0031
0	0.3547	0.0284	-0.1246	0.0849	0.0046	0	0.2121	0.0423	-0.1400	0.0276	0.0040
1	0.4293	0.0346	-0.1230	0.0519	0.0063	1	0.3159	0.0467	-0.1517	0.0390	0.0052
2	0.5040	0.0431	-0.1156	0.0602	0.0072	2	0.4272	0.0561	-0.1626	0.0516	0.0071
3	0.5854	0.0539	-0.1104	0.0696	0.0105	3	0.5132	0.0679	-0.1688	0.0615	0.0088
5	0.7520	0.0820	-0.0964	0.0896	0.0079	5	0.6748	0.0996	-0.1826	0.0810	0.0137
7	0.8684	0.1210	-0.0970	0.1070	0.0100						
10	1.0079	0.1878	-0.1047	0.1255	0.0293						
15	0.9536	0.1434	-0.1511	0.1176	0.0401						
20	0.9575	0.1921	-0.1486	0.1200	0.0520						
25	0.9420	0.2393	-0.1407	0.1184	0.0631						
$M = 0.90$											
-10	-0.4911	0.0570	-0.0867	-0.0606	0.0092	-10	-0.5634	0.1301	-0.0095	-0.0657	0.0121
-7	-0.2895	0.0307	-0.1176	-0.0329	0.0051	-7	-0.3665	0.0809	-0.0504	-0.0407	0.0065
-5	-0.1164	0.0175	-0.1325	-0.0107	0.0024	-5	-0.2185	0.0576	-0.0762	-0.0227	0.0039
-3	0.0565	0.0128	-0.1473	0.0093	0.0019	-3	-0.0819	0.0446	-0.0958	-0.0052	0.0027
-2	0.1591	0.0128	-0.1485	0.0201	0.0022	-2	0.0144	0.0417	-0.1093	0.0052	0.0026
-1	0.2533	0.0143	-0.1517	0.0305	0.0029	-1	0.0934	0.0403	-0.1205	0.0140	0.0031
0	0.3422	0.0169	-0.1480	0.0415	0.0040	0	0.1983	0.0435	-0.1330	0.0256	0.0037
1	0.4261	0.0202	-0.1454	0.0519	0.0055	1	0.2989	0.0474	-0.1465	0.0369	0.0050
2	0.5442	0.0261	-0.1466	0.0640	0.0080	2	0.3967	0.0562	-0.1528	0.0480	0.0064
3	0.6331	0.0329	-0.1452	0.0758	0.0104	3	0.4858	0.0682	-0.1627	0.0590	0.0082
5	0.7803	0.0480	-0.1490	0.0941	0.0155	5	0.6467	0.1018	-0.1752	0.0779	0.0128
7	0.9241	0.0690	-0.1566	0.1101	0.0219						
$M = 1.10$											
-10	-0.4911	0.0570	-0.0867	-0.0606	0.0092	-10	-0.5634	0.1301	-0.0095	-0.0657	0.0121
-7	-0.2895	0.0307	-0.1176	-0.0329	0.0051	-7	-0.3665	0.0809	-0.0504	-0.0407	0.0065
-5	-0.1164	0.0175	-0.1325	-0.0107	0.0024	-5	-0.2185	0.0576	-0.0762	-0.0227	0.0039
-3	0.0565	0.0128	-0.1473	0.0093	0.0019	-3	-0.0819	0.0446	-0.0958	-0.0052	0.0027
-2	0.1591	0.0128	-0.1485	0.0201	0.0022	-2	0.0144	0.0417	-0.1093	0.0052	0.0026
-1	0.2533	0.0143	-0.1517	0.0305	0.0029	-1	0.0934	0.0403	-0.1205	0.0140	0.0031
0	0.3422	0.0169	-0.1480	0.0415	0.0040	0	0.1983	0.0435	-0.1330	0.0256	0.0037
1	0.4261	0.0202	-0.1454	0.0519	0.0055	1	0.2989	0.0474	-0.1465	0.0369	0.0050
2	0.5442	0.0261	-0.1466	0.0640	0.0080	2	0.3967	0.0562	-0.1528	0.0480	0.0064
3	0.6331	0.0329	-0.1452	0.0758	0.0104	3	0.4858	0.0682	-0.1627	0.0590	0.0082
5	0.7803	0.0480	-0.1490	0.0941	0.0155	5	0.6467	0.1018	-0.1752	0.0779	0.0128
7	0.9241	0.0690	-0.1566	0.1101	0.0219						

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{c_t}{c} = 0.30$$

$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$	$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$
$M = 0.40$										$M = 0.95$	
-10	-2534					-10	-4754	.1149	-0.0709	-0.0536	.0155
-7	-0.0576					-7	-2659	.0682	-0.1084	-0.0259	.0097
-5	.0504					-5	-0.0725	.0444	-0.1440	-0.0031	.0070
-3	.1656					-3	.1128	.0376	-0.1702	.0176	.0064
-2	.2232					-2	.2176	.0388	-0.1802	.0287	.0068
-1	.2750					-1	.3223	.0428	-0.1806	.0403	.0074
0	.3441					0	.4077	.0508	-0.1826	.0499	.0086
1	.3888					1	.4899	.0587	-0.1820	.0593	.0102
2	.4391					2	.5882	.0713	-0.1823	.0694	.0124
3	.4895					3	.6607	.0864	-0.1830	.0792	.0150
5	.6119					5	.8154	.1240	-0.1937	.0981	.0205
7	.7343										
10	.8207										
15	.8639										
20	.8236										
25	.7861										
$M = 0.60$										$M = 1.00$	
-10	-2827	.0444	-0.0774	-0.0323	.0084	-10	-5014	.1203	-0.0443	-0.0557	.0152
-7	-0.0705	.0098	-0.1075	-0.0057	.0046	-7	-2959	.0743	-0.0854	-0.0296	.0098
-5	.0677	.0036	-0.1046	.0107	.0034	-5	-1288	.0531	-0.1164	-0.0099	.0076
-3	.1901	.0063	-0.1049	.0249	.0039	-3	.0414	.0434	-0.1454	.0095	.0066
-2	.2635	.0115	-0.0984	.0330	.0048	-2	.1457	.0422	-0.1602	.0211	.0067
-1	.3255	.0168	-0.0962	.0408	.0056	-1	.2683	.0441	-0.1741	.0346	.0070
0	.4059	.0245	-0.0948	.0485	.0061	0	.3649	.0498	-0.1815	.0454	.0076
1	.4515	.0315	-0.0958	.0556	.0072	1	.4492	.0584	-0.1817	.0552	.0089
2	.5156	.0403	-0.0871	.0628	.0089	2	.5412	.0705	-0.1816	.0642	.0110
3	.5726	.0508	-0.0822	.0707	.0108	3	.6164	.0863	-0.1830	.0738	.0135
5	.7264	.0833	-0.0716	.0896	.0320	5	.7819	.1237	-0.1987	.0927	.0180
7	.8261	.1208	-0.0693	.1040	.0216						
10	.9059	.1926	-0.1007	.1173	.0311						
15	.9201	.2998	-0.1384	.1196	.0436						
20	.8745	.3894	-0.1375	.1141	.0527						
25	.8631	.4736	-0.1404	.0838	.0631						
$M = 0.80$										$M = 1.05$	
-10	-3048	.0618	-0.0951	-0.0342	.0091	-10	-4888	.1158	-0.0397	-0.0550	.0134
-7	-0.0968	.0238	-0.1348	-0.0080	.0049	-7	-2948	.0729	-0.0815	-0.0301	.0085
-5	.0629	.0143	-0.1347	.0099	.0041	-5	-1407	.0535	-0.1083	-0.0111	.0135
-3	.1935	.0155	-0.1342	.0255	.0043	-3	.0222	.0437	-0.1331	.0072	.0058
-2	.2854	.0202	-0.1316	.0349	.0049	-2	.1155	.0437	-0.1464	.0174	.0057
-1	.3454	.0254	-0.1306	.0427	.0056	-1	.2118	.0455	-0.1625	.0286	.0061
0	.4189	.0321	-0.1251	.0513	.0070	0	.3333	.0503	-0.1751	.0422	.0069
1	.5015	.0411	-0.1223	.0609	.0088	1	.4177	.0583	-0.1809	.0518	.0078
2	.5689	.0497	-0.1131	.0685	.0104	2	.5110	.0710	-0.1795	.0616	.0094
3	.6386	.0613	-0.1083	.0775	.0063	3	.5895	.0856	-0.1843	.0713	.0117
5	.7856	.0935	-0.1003	.0951	.0092	5	.7406	.1221	-0.1945	.0887	.0164
7	.8804	.1342	-0.0990	.1119	.0058	7	.8591	.1654	-0.2104	.1031	.0221
10	1.0178	.2022	-0.1032	.1296	.0082						
$M = 0.90$										$M = 1.10$	
-10	-3457	.0800	-0.1157	-0.0383	.0115	-10	-4810	.1137	-0.0355	-0.0542	.0119
-7	-0.1494	.0440	-0.1424	-0.0133	.0067	-7	-2900	.0706	-0.0779	-0.0298	.0072
-5	.0171	.0279	-0.1538	.0057	.0051	-5	-1393	.0519	-0.1044	-0.0115	.0052
-3	.1750	.0273	-0.1627	.0234	.0051	-3	.0144	.0441	-0.1272	.0060	.0048
-2	.2604	.0283	-0.1522	.0320	.0055	-2	.1034	.0445	-0.1398	.0160	.0048
-1	.3560	.0367	-0.1634	.0428	.0070	-1	.2010	.0466	-0.1525	.0263	.0052
0	.4097	.0410	-0.1510	.0506	.0077	0	.2972	.0508	-0.1660	.0376	.0062
1	.5019	.0493	-0.1507	.0610	.0097	1	.3977	.0572	-0.1762	.0495	.0075
2	.5822	.0588	-0.1379	.0701	.0120	2	.4810	.0698	-0.1755	.0582	.0085
3	.7529	.0709	-0.1610	.0805	.0142	3	.5657	.0858	-0.1773	.0680	.0102
5	.7853	.1018	-0.1388	.0970	.0193	5	.7093	.1235	-0.1916	.0860	.0140
7	.9151	.1444	-0.0969	.1115	.0255	7	.8327	.1684	-0.1990	.0999	.0193

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.55$											
-10	-0.1961					-10	-0.3709	.1138	-0.1084	-0.0396	.0130
-7	-0.0087					-7	-0.1516	.0761	-0.1425	-0.0119	.0084
-5	.1096					-5	.0468	.0603	-0.1726	.0101	.0063
-3	.2076					-3	.2242	.0575	-0.1894	.0297	.0065
-2	.2711					-2	.3322	.0603	-0.1913	.0411	.0072
-1	.3157					-1	.4177	.0670	-0.1875	.0509	.0084
0	.3806					0	.4935	.0761	-0.1876	.0524	.0098
1	.4239					1	.5822	.0860	-0.1856	.0693	.0118
2	.4844					2	.6563	.0992	-0.1845	.0789	.0140
3	.5306					3	.7321	.1158	-0.1838	.0897	.0166
5	.6546					5	.8708	.1526	-0.1898	.1060	.0227
7	.7584										
10	.8276										
15	.8333										
20	.8045										
25	.7757										
$M = 0.60$											
$M = 1.00$											
-10	-0.2161	.0534	-0.0830	-0.0239	.0061	-10	-0.4235	.1188	-0.0725	-0.0447	.0129
-7	-0.0078	.0263	-0.1053	.0021	.0027	-7	-0.1995	.0800	-0.1153	-0.0175	.0086
-5	.1405	.0263	-0.0983	.0190	.0022	-5	-0.0230	.0630	-0.1452	.0026	.0068
-3	.2739	.0312	-0.0972	.0341	.0034	-3	.1596	.0592	-0.1735	.0228	.0062
-2	.3417	.0368	-0.0906	.0429	.0040	-2	.2854	.0596	-0.1849	.0360	.0067
-1	.4052	.0452	-0.0900	.0501	.0045	-1	.3744	.0653	-0.1862	.0461	.0075
0	.4651	.0544	-0.0902	.0573	.0055	0	.4587	.0743	-0.1864	.0562	.0090
1	.5265	.0639	-0.0869	.0651	.0071	1	.5401	.0856	-0.1859	.0652	.0103
2	.5864	.0750	-0.0766	.1440	.0109	2	.6198	.0985	-0.1859	.0743	.0126
3	.6506	.0885	-0.0772	.0814	.0109	3	.6981	.0970	-0.1904	.0857	.0151
5	.7676	.1176	-0.0651	.0961	.0158	5	.8469	.1554	-0.2046	.1015	.0103
7	.8646	.1621	-0.0711	.1111	.0220						
10	.9017	.2280	-0.1021	.1189	.0310						
15	.8960	.3241	-0.1344	.1160	.0209						
20	.8674	.4126	-0.1369	.1134	.0260						
25	.8674	.5081	-0.1413	.1131	.0314						
$M = 0.80$											
$M = 1.05$											
-10	-0.2363	.0671	-0.0926	-0.0257	.0077	-10	-0.4119	.1148	-0.0693	-0.0439	.0106
-7	-0.0107	.0350	-0.1323	-0.0018	.0038	-7	-0.2015	.0783	-0.1090	-0.0183	.0087
-5	.1491	.0326	-0.1323	.0200	.0034	-5	-0.0385	.0638	-0.1353	.0003	.0078
-3	.2915	.0390	-0.1279	.0362	.0042	-3	.1171	.0601	-0.1591	.0183	.0073
-2	.3622	.0450	-0.1244	.0447	.0051	-2	.2504	.0601	-0.1763	.0325	.0074
-1	.4300	.0514	-0.1228	.0523	.0064	-1	.3467	.0648	-0.1822	.0430	.0078
0	.4920	.0596	-0.1225	.0605	.0074	0	.4282	.0739	-0.1857	.0527	.0086
1	.5559	.0695	-0.1146	.0690	.0093	1	.5097	.0856	-0.1839	.0619	.0093
2	.6315	.0793	-0.1070	.0768	.0111	2	.5912	.0988	-0.1851	.0715	.0108
3	.6993	.0922	-0.1029	.0856	.0135	3	.6579	.1136	-0.1891	.0812	.0128
5	.8252	.1272	-0.0953	.1046	.0191	5	.8179	.1184	-0.2018	.0980	.0187
7	.9123	.1653	-0.0900	.1171	.0238						
10	1.0034	.2344	-0.0907	.1301	.0334						
$M = 0.90$											
$M = 1.10$											
-10	-0.2701	.0921	-0.1347	-0.0285	.0104	-10	-0.3963	.1101	-0.0668	-0.0433	.0100
-7	-0.0436	.0576	-0.1665	-0.0009	.0063	-7	-0.2024	.0773	-0.1042	-0.0193	.0058
-5	.1333	.0491	-0.1805	.0192	.0053	-5	-0.0488	.0643	-0.1290	-0.0016	.0047
-3	.2855	.0513	-0.1812	.0360	.0061	-3	.1091	.0607	-0.1531	.0173	.0047
-2	.3752	.0585	-0.1802	.0462	.0073	-2	.2039	.0625	-0.1632	.0279	.0050
-1	.4513	.0651	-0.1725	.0550	.0085	-1	.3144	.0678	-0.1740	.0399	.0056
0	.5334	.0765	-0.1801	.0652	.0102	0	.4078	.0759	-0.1809	.0504	.0064
1	.5932	.0849	-0.1695	.0730	.0118	1	.4867	.0865	-0.1812	.0598	.0079
2	.6650	.0933	-0.1523	.0811	.0139	2	.5657	.1006	-0.1815	.0681	.0095
3	.7282	.1047	-0.1459	.0906	.0162	3	.6231	.1172	-0.1853	.0816	.0115
5	.8513	.1396	-0.1484	.1062	.0214	5	.7868	.1246	-0.1963	.0950	.0166

TABLE 3 .- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Continued

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.5310					-10	-0.7168	0.1453	0.0848	-0.0887	-0.0220
-7	-0.3947					-7	-0.5165	0.0850	0.0382	-0.0631	-0.0121
-5	-0.2942					-5	-0.3900	0.0569	0.0173	-0.0461	-0.0077
-3	-0.1973					-3	-0.2382	0.0363	0.0000	-0.0273	-0.0037
-2	-0.1471					-2	-0.1644	0.0270	-0.0037	-0.0188	-0.0023
-1	-0.0969					-1	-0.0949	0.0249	-0.0028	-0.0102	-0.0015
0	-0.0466					0	-0.0148	0.0238	0.0009	-0.0007	-0.0012
1	0.0179					1	0.0632	0.0259	0.0033	0.0085	-0.0012
2	0.0789					2	0.1328	0.0280	0.0019	0.0171	-0.0015
3	0.1327					3	0.2087	0.0342	-0.0014	0.0256	-0.0020
5	0.2404					5	0.3373	0.0519	-0.0107	0.0416	-0.0039
7	0.3588					7	0.4891	0.0829	0.0411	0.0604	-0.0076
10	0.5238					10	0.6852	0.1453	-0.0877	0.0863	-0.0156
15	0.6925					15	0.9593	0.2800	-0.1483	0.1218	-0.0341
20	0.7140					20	0.9804	0.3940	-0.2043	0.1218	-0.0482
25	0.6996					25	1.0225	0.5226	-0.2453	0.1262	-0.0644
$M = 0.60$											
-10	-0.6001	0.1219	0.0261	-0.0741	-0.0163	-10	-0.7372	0.1620	0.1193	-0.0866	-0.0229
-7	-0.4376	0.0598	0.0016	-0.0520	-0.0077	-7	-0.5453	0.0994	0.0733	-0.0631	-0.0137
-5	-0.3250	0.0329	-0.0123	-0.0371	-0.0036	-5	-0.4141	0.0705	0.0496	-0.0467	-0.0097
-3	-0.2068	0.0218	-0.0090	-0.0227	-0.0019	-3	-0.2626	0.0487	0.0286	-0.0301	-0.0061
-2	-0.1293	0.0218	0.0025	-0.0149	-0.0009	-2	-0.1777	0.0426	0.0187	-0.0203	-0.0042
-1	-0.0776	0.0144	0.0041	-0.0081	-0.0004	-1	-0.1030	0.0347	0.0099	-0.0114	-0.0027
0	-0.0203	0.0126	-0.0097	-0.0036	-0.0003	0	-0.0141	0.0347	0.0053	-0.0016	-0.0023
1	0.0480	0.0144	-0.0130	0.0066	0.0004	1	0.0707	0.0378	-0.0036	0.0082	-0.0024
2	0.1052	0.0162	-0.0130	0.0131	0.0012	2	0.1495	0.0418	0.0103	0.0180	-0.0030
3	0.1699	0.0218	-0.0148	0.0203	0.0009	3	0.2323	0.0477	-0.0214	0.0265	-0.0035
5	0.2880	0.0329	-0.0171	0.0341	-0.0002	5	0.3737	0.0667	0.0402	0.0435	-0.0055
7	0.4210	0.0510	-0.0196	0.0505	-0.0021	7	0.5191	0.0974	-0.0697	0.0611	-0.0090
10	0.5964	0.1034	-0.0359	0.0729	-0.0079	10	0.7231	0.1620	-0.1220	0.0856	-0.0168
15	0.7349	0.2197	-0.1152	0.0944	-0.0220	15	1.0180	0.3050	-0.1934	0.1245	-0.0359
20	0.7515	0.3143	-0.1527	0.0962	-0.0351	20	1.2119	0.4688	-0.2350	0.1497	-0.0584
25	0.7478	0.3996	-0.1634	0.0938	-0.0454	25	1.2200	0.6158	-0.2931	0.1497	-0.0384
$M = 0.80$											
-10	-0.6504	0.1156	0.0160	-0.0795	-0.0171	-10	-0.7197	0.1573	0.1189	-0.0851	-0.0228
-7	-0.5053	0.0640	-0.0061	-0.0593	-0.0096	-7	-0.5374	0.0964	0.0760	-0.0618	-0.0142
-5	-0.3627	0.0345	-0.0144	-0.0417	-0.0066	-5	-0.4016	0.0677	0.0545	-0.0461	-0.0101
-3	-0.2201	0.0185	-0.0127	-0.0245	-0.0021	-3	-0.2503	0.0477	0.0322	-0.0289	-0.0056
-2	-0.1551	0.0160	-0.0088	-0.0168	-0.0014	-2	-0.1707	0.0429	0.0223	-0.0195	-0.0045
-1	-0.0926	0.0148	-0.0055	-0.0099	-0.0009	-1	-0.0970	0.0382	0.0129	-0.0116	-0.0034
0	-0.0150	0.0135	0.0000	-0.0008	-0.0009	0	-0.0136	0.0372	0.0047	-0.0025	-0.0026
1	0.0575	0.0160	0.0006	0.0079	-0.0006	1	0.0640	0.0382	-0.0034	0.0072	-0.0026
2	0.1201	0.0173	0.0022	0.0152	-0.0006	2	0.1416	0.0429	-0.0129	0.0157	-0.0029
3	0.1926	0.0223	0.0066	0.0233	-0.0010	3	0.2231	0.0497	-0.0227	0.0248	-0.0037
5	0.3252	0.0345	0.0061	0.0395	-0.0020	5	0.3589	0.0677	-0.0434	0.0408	-0.0056
7	0.4778	0.0615	-0.0044	0.0577	-0.0048	7	0.5083	0.1001	-0.0721	0.0587	-0.0091
10	0.6429	0.1143	-0.0265	0.0789	-0.0110	10	0.6887	0.1573	-0.1184	0.0807	-0.0165
15	0.7505	0.2251	-0.1113	0.0963	-0.0257	15	1.0243	0.3053	-0.2017	0.1218	-0.0360
20	0.7830	0.3200	-0.1555	0.0988	-0.0387	20	1.2649	0.4980	-0.2686	0.1532	-0.0611
25	0.8055	0.4158	-0.1704	0.1012	-0.0514	25	1.3308	0.6679	-0.3004	0.1664	-0.0839
$M = 0.90$											
-10	-0.6870	0.1330	0.0500	-0.0864	-0.0202	-10	-0.7087	0.1537	0.1172	-0.0833	-0.0245
-7	-0.5186	0.0740	0.0172	-0.0628	-0.0109	-7	-0.5259	0.0949	0.0750	-0.0616	-0.0147
-5	-0.3989	0.0437	0.0010	-0.061	-0.0066	-5	-0.3916	0.0651	0.0545	-0.0459	-0.0097
-3	-0.2549	0.0239	-0.0064	-0.0287	-0.0031	-3	-0.2424	0.0459	0.0322	-0.0287	-0.0062
-2	-0.1773	0.0197	-0.0064	0.0038	-0.0021	-2	-0.1623	0.0390	0.0215	-0.0196	-0.0050
-1	-0.1045	0.0175	-0.0039	-0.0113	-0.0014	-1	-0.0877	0.0352	0.0136	-0.0115	-0.0036
0	-0.0222	0.0164	-0.0015	-0.0018	-0.0009	0	-0.0093	0.0343	0.0046	-0.0027	-0.0028
1	0.0621	0.0164	-0.0015	0.0081	-0.0008	1	0.0746	0.0358	-0.0054	0.0069	-0.0029
2	0.1374	0.0197	0.0035	0.0170	-0.0009	2	0.1492	0.0399	-0.0148	0.0151	-0.0033
3	0.2172	0.0262	0.0035	0.0256	-0.0012	3	0.2294	0.0459	-0.0231	0.0235	-0.0038
5	0.3435	0.0414	-0.0029	0.0414	-0.0029	5	0.3637	0.0651	-0.0454	0.0395	-0.0060
7	0.4876	0.0709	-0.0225	0.0595	-0.0062	7	0.5073	0.0953	-0.0672	0.0570	-0.0098
10	0.6649	0.1308	-0.0554	0.0843	-0.0137	10	0.6900	0.1537	-0.1147	0.0791	-0.0169
15	0.8865	0.2529	-0.1010	0.1140	-0.0302	15	0.9996	0.2980	-0.2005	0.1177	-0.0356
20	0.8754	0.3542	-0.1804	0.1079	-0.0215	20	1.2421	0.4897	-0.2624	0.1479	-0.0606
25	0.9175	0.4665	-0.2025	0.1130	-0.0291	25	1.3801	0.6953	-0.3160	0.1690	-0.0870
$M = 1.10$											
-10	-0.7168	0.1453	0.0848	-0.0887	-0.0220	-10	-0.750	0.0750	-0.0616	-0.0147	
-7	-0.5165	0.0850	0.0382	-0.0631	-0.0121	-7	-0.775	0.0950	-0.0750	-0.0147	
-5	-0.3900	0.0569	0.0173	-0.0461	-0.0077	-5	-0.800	0.0650	-0.0550	-0.0147	
-3	-0.2382	0.0363	0.0000	-0.0273	-0.0037	-3	-0.825	0.0450	-0.0350	-0.0147	
-2	-0.1644	0.0270	-0.0037	-0.0188	-0.0023	-2	-0.850	0.0250	-0.0150	-0.0147	
-1	-0.0949	0.0249	-0.0028	-0.0102	-0.0015	-1	-0.875	0.0150	-0.0050	-0.0147	
0	-0.0148	0.0238	0.0009	-0.0007	-0.0012	0	-0.900	0.0050	-0.0000	-0.0147	
1	0.0632	0.0259	0.0033	0.0085	-0.0012	1	0.925	0.0250	-0.0150	-0.0147	
2	0.1328	0.0280	0.0019	0.0171	-0.0012	2	0.950	0.0450	-0.0250	-0.0147	
3	0.2087	0.0342	-0.0014	0.0256	-0.0020	3	0.975	0.0650	-0.0450	-0.0147	
5	0.3373	0.0519	-0.0107	0.0416	-0.0039	5	1.000	0.0850	-0.0650	-0.0147	
7	0.4891	0.0829	0.0411	0.0604	-0.0076	7	1.025	0.1150	-0.0850	-0.0147	
10	0.6852	0.1453	-0.0877	0.0863	-0.0156	10	1.050	0.1450	-0.1050	-0.0147	
15	0.9593	0.2800	-0.1483	0.1218	-0.0341	15	1.075	0.2150	-0.1250	-0.0147	
20	0.9804	0.3940	-0.2043	0.1218	-0.0482	20	1.100	0.2450	-0.1450	-0.0147	
25	1.0225	0.5226	-0.2453	0.1262	-0.0644	25	1.125	0.3650	-0.1650	-0.0147	

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.20$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.2415					-10	-0.5179	0.1279	-0.0211	-0.0634	-0.0148
-7	-0.0972					-7	-0.3615	0.0801	-0.0444	-0.0412	-0.0082
-5	0.0252					-5	-0.2114	0.0552	-0.0748	-0.0224	-0.0037
-3	0.1512					-3	-0.0317	0.0395	-0.1099	-0.0019	-0.0008
-2	0.1908					-2	0.0740	0.0395	-0.1239	0.0101	-0.0003
-1	0.2520					-1	0.1797	0.0395	-0.1426	0.0226	-0.0002
0	0.3060					0	0.2790	0.0446	-0.1519	0.0334	-0.0010
1	0.3600					1	0.3699	0.0552	-0.1637	0.0436	-0.0020
2	0.4104					2	0.4545	0.0634	-0.1637	0.0534	-0.0032
3	0.4680					3	0.5390	0.0738	-0.1730	0.0640	-0.0046
5	0.5508					5	0.6743	0.0520	-0.1824	0.0636	-0.0082
7	0.6696					7	0.8223	0.1518	-0.2105	0.0947	-0.144
10	0.8208					10	0.9978	0.2266	-0.2450	0.1201	-0.243
15	0.9071					15	1.2134	0.3670	-0.2665	0.1485	-0.425
20	0.8783					20	1.0908	0.4501	-0.2684	0.1279	-0.0518
25	0.8424					25	1.0781	0.5676	-0.2759	0.1259	-0.0659
$M = 0.60$											
$M = 1.00$											
-10	-0.2777	0.0800	-0.0983	-0.0324	-0.0036	-10	-0.5509	0.1385	-0.0067	-0.0672	-0.0160
-7	-0.1055	0.0337	-0.1229	-0.0093	0.0010	-7	-0.3605	0.0907	-0.0381	-0.0429	-0.0087
-5	0.0167	0.0237	-0.1246	-0.0042	0.0024	-5	-0.2086	0.0676	-0.0694	-0.0251	-0.0053
-3	0.1444	0.0211	-0.1253	-0.0186	0.0032	-3	-0.0608	0.0557	-0.0919	-0.0084	-0.0031
-2	0.2037	0.0211	-0.1246	-0.0252	0.0035	-2	0.0223	0.0529	-0.1066	0.0015	-0.0024
-1	0.2610	0.0255	-0.1229	-0.0312	0.0030	-1	0.1154	0.0518	-0.1223	0.0116	-0.0021
0	0.3240	0.0300	-0.1212	-0.0387	0.0023	0	0.2309	0.0557	-0.1388	0.0233	-0.0022
1	0.4036	0.0344	-0.1212	-0.0473	0.0020	1	0.3220	0.0628	-0.1532	0.0342	-0.0028
2	0.4666	0.0437	-0.1212	-0.0545	0.0017	2	0.4253	0.0717	-0.1702	0.0457	-0.0039
3	0.5091	0.0481	-0.1146	-0.0602	0.0014	3	0.5124	0.0857	-0.1859	0.0567	-0.0055
5	0.6258	0.0674	-0.1106	-0.0740	-0.0007	5	0.6622	0.1146	-0.2074	0.0744	-0.0091
7	0.7461	0.1011	-0.1130	-0.0896	-0.0046	7	0.8162	0.1584	-0.2311	0.0924	-0.148
10	0.8757	0.1692	-0.1351	-0.1073	-0.0122	10	0.9924	0.2351	-0.2598	0.1137	-0.243
15	0.9201	0.2822	-0.1965	-0.1153	-0.0267	15	1.2637	0.2939	-0.3046	0.1455	-0.444
20	0.8961	0.3744	-0.2137	-0.1105	-0.0389	20	1.3974	0.2839	-0.3180	0.1665	-0.1350
25	0.8720	0.4562	-0.2154	-0.1061	-0.0492	25	1.3447	0.3516	-0.3566	0.1566	-0.0813
$M = 0.80$											
$M = 1.05$											
-10	-0.3361	0.0758	-0.1009	-0.0398	-0.0068	-10	-0.5464	0.1377	-0.0121	-0.0719	-0.0162
-7	-0.1768	0.0381	-0.1248	-0.0187	-0.0017	-7	-0.3578	0.0908	-0.0323	-0.0497	-0.0091
-5	-0.0439	0.0233	-0.1292	-0.0030	0.0009	-5	-0.2178	0.0698	-0.0524	-0.0332	-0.0057
-3	0.1041	0.0216	-0.1315	-0.0134	0.0015	-3	-0.0603	0.0554	-0.0838	-0.0154	-0.0029
-2	0.1819	0.0173	-0.1304	-0.0221	0.0015	-2	0.0117	0.0535	-0.0968	-0.0071	-0.0026
-1	0.2546	0.0266	-0.1304	-0.0302	0.0014	-1	0.1050	0.0535	-0.1110	0.0032	-0.0023
0	0.3198	0.0314	-0.1298	-0.0386	0.0010	0	0.2003	0.0554	-0.1239	0.0135	-0.0023
1	0.4013	0.0389	-0.1331	-0.0481	0.0005	1	0.2820	0.0622	-0.1368	0.0230	-0.0032
2	0.4803	0.0469	-0.1359	-0.0578	-0.0002	2	0.3733	0.0698	-0.1514	0.0332	-0.0042
3	0.5456	0.0554	-0.1292	-0.0656	-0.0012	3	0.4628	0.0813	-0.1669	0.0431	-0.0057
5	0.6823	0.0790	-0.1248	-0.0808	-0.0043	5	0.6320	0.1118	-0.1987	0.0628	-0.0246
7	0.8027	0.1179	-0.1304	-0.0962	-0.0087	7	0.7778	0.1540	-0.2237	0.0806	-0.0149
10	0.9256	0.1776	-0.1399	-0.1136	-0.0168	10	0.9606	0.2295	-0.2564	0.1019	-0.245
15	0.9230	0.2824	-0.2009	-0.1132	-0.0310	15	1.2328	0.3903	-0.3071	0.1337	-0.445
20	0.9406	0.3873	-0.2219	-0.1132	-0.0476	20	1.4273	0.5890	-0.3484	0.1598	-0.678
25	0.9356	0.4811	-0.2275	-0.1120	-0.0552	25	1.4195	0.7364	-0.3613	0.1586	-0.0889
$M = 0.90$											
$M = 1.10$											
-10	-0.4663	0.1055	-0.0565	-0.0559	-0.0120	-10	-0.5311	0.1361	-0.0116	-0.0737	-0.0167
-7	-0.2920	0.0606	-0.0835	-0.0325	-0.0049	-7	-0.3459	0.0883	-0.0310	-0.0523	-0.0104
-5	-0.1144	0.0344	-0.1228	-0.0106	0.0004	-5	-0.2113	0.0671	-0.0571	-0.0365	-0.0076
-3	0.0477	0.0278	-0.1375	-0.0077	0.0008	-3	-0.0636	0.0542	-0.0815	-0.0198	-0.0044
-2	0.1443	0.0284	-0.1425	-0.0190	0.0009	-2	0.0224	0.0524	-0.0951	-0.0103	-0.0036
-1	0.2243	0.0318	-0.1425	-0.0279	0.0005	-1	0.1028	0.0524	-0.1138	-0.0015	-0.0032
0	0.2976	0.0389	-0.1449	-0.0455	0.0001	0	0.1870	0.0542	-0.1195	0.0079	-0.0033
1	0.3775	0.0449	-0.1449	-0.0451	-0.0005	1	0.2767	0.0598	-0.1332	0.0180	-0.0040
2	0.4563	0.0535	-0.1483	-0.0555	-0.0016	2	0.3590	0.0681	-0.1460	0.0274	-0.0046
3	0.5529	0.0644	-0.1522	-0.0672	-0.0034	3	0.4469	0.0791	-0.1605	0.0369	-0.0062
5	0.6884	0.0917	-0.1596	-0.0825	-0.0065	5	0.5946	0.1075	-0.1882	0.0543	-0.0096
7	0.8061	0.1310	-0.1695	-0.0985	-0.0116	7	0.7517	0.1481	-0.2151	0.0728	-0.149
10	0.9438	0.1972	-0.1866	-0.1182	-0.0208	10	0.9256	0.2216	-0.2482	0.0941	-0.241
15	1.1414	0.3309	-0.2137	-0.1416	-0.0396	15	1.1967	0.3779	-0.2978	0.1253	-0.431
20	1.0437	0.4248	-0.2555	-0.1236	-0.0491	20	1.3837	0.5720	-0.3408	0.1492	-0.664
25	1.0437	0.5330	-0.2604	-0.1232	-0.0630	25	1.4885	0.7781	-0.3830	0.1646	-0.0936

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
M = 0.40											
M = 0.55											
-10	-0.1905					-10	-0.4415	0.1185	-0.0739	-0.0537	-0.0160
-7	-0.0395					-7	-0.2884	0.0744	-0.0911	-0.0313	-0.0095
-5	-0.0898					-5	-0.1447	0.0534	-0.1191	-0.0115	-0.0056
-3	-0.1977					-3	-0.0866	0.0456	-0.1542	0.0125	-0.0034
-2	-0.2444					-2	-0.1996	0.0484	-0.1705	0.0253	-0.0038
-1	-0.3055					-1	-0.2873	0.0613	-0.1916	0.0355	-0.0050
0	-0.3738					0	-0.3876	0.0872	-0.1869	0.0467	-0.0053
1	-0.4169					1	-0.4774	0.0744	-0.1986	0.0578	-0.0071
2	-0.4852					2	-0.5598	0.0847	-0.1963	0.0667	-0.0084
3	-0.5391					3	-0.6232	0.0925	-0.1893	0.0747	-0.0093
5	-0.6325					5	-0.7647	0.1314	-0.2103	0.0923	-0.0139
7	-0.7475					7	-0.8978	0.1756	-0.2173	0.1094	-0.0197
10	-0.8769					10	1.0605	0.2535	-0.2477	0.1299	-0.0304
15	-0.9128					15	1.2126	0.3927	-0.2477	0.1521	-0.0246
20	-0.8697					20	1.1492	0.5017	-0.2757	0.1378	-0.0308
25	-0.8194					25	1.1070	0.6088	-0.2757	0.1309	-0.0375
M = 0.60											
M = 1.00											
-10	-0.2312	0.0629	-0.0966	-0.0287	-0.0082	-10	-0.5058	0.1333	-0.0403	-0.0575	-0.0167
-7	-0.0518	0.0237	-0.1228	-0.0045	-0.0023	-7	-0.2934	0.0886	-0.0805	-0.0319	-0.0100
-5	-0.0869	0.0163	-0.1228	0.0114	-0.0014	-5	-0.1315	0.0596	-0.1119	-0.0124	-0.0063
-3	-0.2201	0.0181	-0.1219	0.0269	-0.0014	-3	-0.0445	0.0587	-0.1410	0.0074	-0.0048
-2	-0.2885	0.0211	-0.1235	0.0344	-0.0016	-2	-0.1558	0.0587	-0.1589	0.0188	-0.0047
-1	-0.3496	0.0255	-0.1210	0.0419	-0.0017	-1	-0.2610	0.0627	-0.1767	0.0316	-0.0049
0	-0.4235	0.0329	-0.1228	0.0494	-0.0023	0	-0.3682	0.0702	-0.1924	0.0434	-0.0056
1	-0.4846	0.0392	-0.1228	0.0569	-0.0026	1	-0.4593	0.0836	-0.2050	0.0540	-0.0071
2	-0.5308	0.0462	-0.1187	0.0625	-0.0032	2	-0.5442	0.0945	-0.2126	0.0635	-0.0084
3	-0.5863	0.0555	-0.1129	0.0688	-0.0044	3	-0.6252	0.1095	-0.2282	0.0732	-0.0102
5	-0.7010	0.0773	-0.1105	0.0829	-0.0069	5	-0.7627	0.1408	-0.2372	0.0900	-0.0143
7	-0.8249	0.1165	-0.1088	0.0988	-0.0116	7	-0.9145	0.1890	-0.2506	0.1080	-0.0209
10	-0.9248	0.1838	-0.1309	0.1134	-0.0204	10	1.0804	0.2717	-0.2730	0.1287	-0.0314
15	-0.9507	0.2985	-0.1801	0.1179	-0.0360	15	1.3151	0.4338	-0.3043	0.1568	-0.0529
20	-0.9137	0.3928	-0.1972	0.1101	-0.0483	20	1.4243	0.6070	-0.3150	0.1748	-0.0765
25	-0.8767	0.4783	-0.2005	0.1048	-0.0585	25	1.3313	0.7283	-0.3357	0.1617	-0.0925
M = 0.80											
M = 1.05											
-10	-0.2795	0.0672	-0.1247	-0.0325	-0.0084	-10	-0.5013	0.1257	-0.0215	-0.0549	-0.0166
-7	-0.0940	0.0308	-0.1442	-0.0091	-0.0035	-7	-0.3031	0.0812	-0.0623	-0.0303	-0.0105
-5	-0.0514	0.0223	-0.1442	0.0077	-0.0017	-5	-0.1516	0.0641	-0.0894	-0.0135	-0.0069
-3	-0.1968	0.0233	-0.1414	0.0243	-0.0017	-3	-0.0136	0.0550	-0.1182	0.0052	-0.0049
-2	-0.2657	0.0266	-0.1414	0.0325	-0.0019	-2	-0.1108	0.0554	-0.1324	0.0159	-0.0046
-1	-0.3346	0.1547	-0.1420	0.0404	-0.0025	-1	-0.1982	0.0583	-0.1461	0.0258	-0.0054
0	-0.4098	0.0634	-0.1425	0.0489	-0.0036	0	-0.2934	0.0645	-0.1612	0.0362	-0.0062
1	-0.4850	0.0481	-0.1414	0.0578	-0.0046	1	-0.3984	0.0731	-0.1762	0.0465	-0.0072
2	-0.5577	0.0574	-0.1414	0.0665	-0.0055	2	-0.4761	0.0841	-0.1891	0.0571	-0.0084
3	-0.6204	0.0667	-0.1319	0.0740	-0.0064	3	-0.5674	0.0985	-0.2012	0.0674	-0.0103
5	-0.7570	0.0942	-0.1286	0.0898	-0.0099	5	-0.7209	0.1329	-0.2218	0.0858	-0.0147
7	-0.8723	0.1374	-0.1319	0.1067	-0.0152	7	-0.8589	0.1792	-0.2390	0.1022	-0.0204
10	-0.9876	0.2010	-0.1375	0.1233	-0.0235	10	1.0377	0.2590	-0.2622	0.1226	-0.0236
15	-0.9400	0.3083	-0.1957	0.1160	-0.0377	15	1.2709	0.4234	-0.3034	0.1522	-0.0529
20	-0.9575	0.4141	-0.2135	0.1148	-0.0513	20	1.4185	0.6117	-0.3352	0.1755	-0.0798
25	-0.9350	0.5141	-0.2190	0.1123	-0.0643	25	1.3874	0.7608	-0.3525	0.1736	-0.0997
M = 0.90											
M = 1.10											
-10	-0.3863	0.0961	-0.0933	-0.0453	-0.0139	-10	-0.4823	0.1279	-0.0198	-0.0537	-0.0173
-7	-0.2087	0.0562	-0.1203	-0.0225	-0.0063	-7	-0.3010	0.0819	-0.0559	-0.0310	-0.0113
-5	-0.0533	0.0371	-0.1448	-0.0031	-0.0032	-5	-0.1552	0.0624	-0.0889	-0.0135	-0.0077
-3	-0.1299	0.0333	-0.1547	0.0171	-0.0024	-3	-0.0150	0.0538	-0.1157	0.0047	-0.0065
-2	-0.2164	0.0360	-0.1547	0.0271	-0.0028	-2	-0.1028	0.0542	-0.1282	0.0145	-0.0062
-1	-0.2908	0.0415	-0.1547	0.0356	-0.0033	-1	-0.1851	0.0565	-0.1406	0.0225	-0.0067
0	-0.3718	0.0480	-0.1562	0.0447	-0.0037	0	-0.2710	0.0611	-0.1546	0.0337	-0.0074
1	-0.4540	0.0590	-0.1596	0.0544	-0.0048	1	-0.3720	0.0703	-0.1675	0.0439	-0.0080
2	-0.5375	0.0682	-0.1596	0.0643	-0.0061	2	-0.4486	0.0809	-0.1790	0.0535	-0.0098
3	-0.6285	0.0830	-0.1611	0.0751	-0.0078	3	-0.5328	0.0948	-0.1910	0.0634	-0.0114
5	-0.7595	0.1134	-0.1669	0.0916	-0.0116	5	-0.6767	0.1279	-0.2133	0.0805	-0.0155
7	-0.8705	0.1534	-0.1719	0.1074	-0.0170	7	-0.8262	0.1738	-0.2299	0.0974	-0.0212
10	-0.9967	0.2200	-0.1792	0.1257	-0.0264	10	-0.9982	0.2537	-0.2547	0.1177	-0.0315
15	1.0966	0.3472	-0.2087	0.1365	-0.0427	15	1.2225	0.4137	-0.2936	0.1455	-0.0519
20	1.0522	0.4564	-0.2430	0.1246	-0.0556	20	1.4020	0.6086	-0.3324	0.1694	-0.0783
25	1.0744	0.5634	-0.2475	0.1218	-0.0692	25	1.4880	0.8128	-0.3672	0.1815	-0.1049

TABLE 3.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 3 MODEL - Concluded

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.55$											
-10	-1438					-10	.3745	.1162	-.0723	-.0408	-.0144
-7	.0108					-7	-.1899	.0757	-.1073	-.0179	-.0088
-5	.1258					-5	-.0179	.0643	-.1446	.0026	-.0065
-3	.2372					-3	.1941	.0622	-.1773	.0261	-.0062
-2	.3127					-2	.2848	.0675	-.1867	.0364	-.0067
-1	.3522					-1	.3692	.0726	-.1890	.0461	-.0074
0	.4133					0	.4557	.0829	-.1937	.0563	-.0089
1	.4528					1	.5338	.0954	-.1960	.0649	-.0106
2	.5103					2	.6118	.1089	-.1983	.0743	-.0124
3	.5571					3	.6899	.1245	-.2030	.0836	-.0144
5	.6613					5	.7954	.1525	-.2030	.0980	-.0183
7	.7799					7	.9135	.1970	-.2170	.1127	-.0237
10	.8913					10	1.0549	.2749	-.2333	.1314	-.0680
15	.8985					15	1.0127	.4150	-.2567	.1536	-.0538
20	.8733					20	1.1181	.5084	-.2754	.1348	-.0640
25	.8158					25	1.0759	.6184	-.2846	.1297	-.0775
$M = 0.60$											
$M = 1.00$											
-10	-1755	.0591	-.0678	-.0209	-.0062	-10	.4142	.1263	-.0482	-.0451	-.0161
-7	.0095	.0255	-.0981	.0030	-.0029	-7	-.2101	.0885	-.0992	-.0203	-.0101
-5	.1478	.0255	-.1021	.0182	-.0019	-5	-.0505	.0249	-.1318	-.0013	-.0072
-3	.2772	.0299	-.1047	.0335	-.0020	-3	.1364	.0695	-.1654	.0191	-.0068
-2	.3381	.0344	-.1021	.0404	-.0024	-2	.2526	.0715	-.1832	.0322	-.0073
-1	.4046	.0410	-.1021	.0478	-.0032	-1	.3485	.0786	-.1945	.0433	-.0086
0	.4601	.0473	-.1021	.0550	-.0036	0	.4223	.0875	-.2007	.0515	-.0089
1	.5229	.0547	-.1021	.0613	-.0042	1	.5092	.0994	-.2056	.0613	-.0103
2	.5691	.0654	-.0998	.0682	-.0047	2	.5859	.1123	-.2109	.0701	-.0117
3	.6356	.0765	-.0981	.0756	-.0057	3	.6668	.1291	-.2154	.0793	-.0141
5	.7484	.1016	-.0956	.0894	-.0087	5	.8082	.1669	-.2324	.0968	-.0188
7	.8595	.1438	-.0981	.1034	-.0139	7	.9193	.2136	-.2369	.1128	-.0249
10	.9608	.2162	-.1348	.1154	-.0232	10	1.0911	.3011	-.2726	.1318	-.0360
15	.9275	.3215	-.1913	.1136	-.0380	15	1.2850	.4570	-.3039	.1576	-.0572
20	.9054	.4161	-.2165	.1085	-.0500	20	1.3861	.6259	-.3263	.1700	-.0790
25	.8795	.5033	-.2281	.1043	-.0602	25	1.2810	.7454	-.3504	.1569	-.0937
$M = 0.80$											
$M = 1.05$											
-10	-2121	.0646	-.0286	-.0237	-.0084	-10	.4075	.1250	-.0428	-.0444	-.0156
-7	.0273	.0363	-.1296	-.0010	-.0040	-7	-.2135	.0887	-.0901	-.0212	-.0099
-5	.1152	.0326	-.1318	.0152	-.0033	-5	-.0640	.0745	-.1223	-.0039	-.0083
-3	.2504	.0326	-.1329	.0312	-.0036	-3	.1106	.0697	-.1524	.0157	-.0071
-2	.3180	.0431	-.1329	.0391	-.0040	-2	.2135	.0716	-.1695	.0272	-.0075
-1	.3881	.0506	-.1340	.0472	-.0048	-1	.3066	.0782	-.1803	.0377	-.0085
0	.4520	.0586	-.1318	.0547	-.0053	0	.3881	.0860	-.1910	.0471	-.0090
1	.5221	.0676	-.1302	.0636	-.0068	1	.4716	.0974	-.1996	.0565	-.0102
2	.5659	.0739	-.1230	.0689	-.0075	2	.5492	.1098	-.2060	.0659	-.0117
3	.6511	.0861	-.1185	.0778	-.0092	3	.6307	.1259	-.2125	.0751	-.0138
5	.7738	.1152	-.1163	.0934	-.0131	5	.7821	.1622	-.2275	.0911	-.0180
7	.8739	.1570	-.1230	.1094	-.0181	7	.8927	.2119	-.2447	.1071	-.0243
10	.9991	.2216	-.1302	.1256	-.0274	10	1.0596	.2940	-.2704	.1266	-.0346
15	.9115	.3203	-.1961	.1114	-.0398	15	1.2730	.4628	-.3176	.1532	-.0564
20	.9365	.4285	-.2216	.1135	-.0534	20	1.4089	.6453	-.3537	.1727	-.0803
25	.9090	.5173	-.2282	.1102	-.0658	25	1.3972	.7922	-.3777	.1715	-.1005
$M = 0.90$											
$M = 1.10$											
-10	-3216	.0976	-.0907	-.0354	-.0123	-10	.3974	.1211	-.0425	-.0433	-.0163
-7	.1297	.0599	-.1227	-.0120	-.0068	-7	-.2183	.0871	-.0866	-.0219	-.0112
-5	.0255	.0490	-.1433	.0063	-.0048	-5	-.0653	.0716	-.1205	-.0041	-.0083
-3	.1940	.0479	-.1497	.0251	-.0047	-3	.1063	.0670	-.1465	.0148	-.0074
-2	.2783	.0534	-.1530	.0350	-.0053	-2	.1959	.0688	-.1630	.0249	-.0076
-1	.3471	.0599	-.1545	.0431	-.0060	-1	.2798	.0670	-.1733	.0340	-.0083
0	.4158	.0654	-.1520	.0513	-.0069	0	.3638	.0826	-.1836	.0438	-.0091
1	.5045	.1353	-.1619	.0619	-.0083	1	.4608	.0918	-.1939	.0543	-.0103
2	.5877	.0883	-.1545	.0709	-.0095	2	.5410	.1056	-.2022	.0635	-.0122
3	.6697	.1020	-.1594	.0816	-.0113	3	.6156	.1201	-.2063	.0720	-.0139
5	.7829	.1326	-.1619	.0969	-.0151	5	.7369	.1541	-.2228	.0872	-.0181
7	.8982	.1772	-.1741	.1127	-.0214	7	.8731	.2047	-.2419	.1035	-.0242
10	1.0201	.2482	-.1864	.1292	-.0310	10	1.0335	.2845	-.2657	.1217	-.0347
15	1.0335	.3524	-.2208	.1256	-.0438	15	1.2499	.4405	-.3054	.1464	-.0548
20	1.0423	.4744	-.2551	.1249	-.0585	20	1.3805	.6423	-.3549	.1672	-.0836
25	1.0113	.5782	-.2620	.1220	-.0715	25	1.4551	.8348	-.3879	.1739	-.1082

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL

 $\frac{t}{c} = 0.04$        $\frac{c_f}{c} = \text{NONE}$ 

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.50$											
-10	-0.6610					-7	-0.7197	0.1094	0.0930	-0.0803	0.0115
-7	-0.5073					-5	-0.5466	0.0623	0.0606	-0.0615	0.0068
-5	-0.3667					-3	-0.3685	0.0325	0.0326	-0.0413	0.0035
-3	-0.2438					-2	-0.2597	0.0192	0.0189	-0.0291	0.0020
-2	-0.1801					-1	-0.1509	0.0137	0.0122	-0.0169	0.0013
-1	-0.1186					0	-0.0346	0.0116	0.0104	-0.0038	0.0011
0	-0.0395					1	0.0866	0.0106	0.0074	0.0105	0.0015
1	0.0285					2	0.2003	0.0152	0.011	0.0229	0.0024
2	0.0878					3	0.2943	0.0228	-0.0088	0.0347	0.0041
3	0.1537					5	0.5095	0.0480	-0.0441	0.0591	0.0085
5	0.2833					7	0.6826	0.0845	-0.0779	0.0782	0.0142
7	0.4282										
10	0.5907										
15	0.6830										
20	0.6896										
25	0.6852										
$M = 0.60$											
$M = 0.80$											
-10	-0.7326	0.1257	0.0284	-0.0844	0.0156	-7	-0.6049	0.0915	0.0729	-0.0642	0.0097
-7	-0.5826	0.0615	-0.0158	-0.0660	0.0083	-5	-0.4112	0.0552	0.0360	-0.0479	0.0057
-5	-0.4152	0.0293	-0.0144	-0.0462	0.0040	-3	-0.2836	0.0343	0.0173	-0.0315	0.0034
3	-0.2717	0.0145	-0.0053	-0.0297	0.0021	-2	-0.1985	0.0265	0.0116	-0.0219	0.0025
-2	-0.1935	0.0107	-0.0027	-0.0204	0.0016	-1	-0.1111	0.0215	0.0097	-0.0118	0.0020
-1	-0.1174	0.0086	0.0041	-0.0115	0.0010	0	-0.0095	0.0195	-0.0009	-0.0004	0.0020
0	-0.0217	0.0064	0.0024	-0.0010	0.0008	1	0.0922	0.0180	-0.0003	0.0115	0.0021
1	0.0543	0.0080	0.0057	0.0089	0.0012	2	0.2056	0.0244	-0.0114	0.0233	0.0034
2	0.1283	0.0117	0.0129	0.0181	0.0018	3	0.2883	0.0297	-0.0195	0.0341	0.0047
3	0.2043	0.0171	0.0179	0.0270	0.0028	5	0.4348	0.0503	-0.0342	0.0505	0.0079
5	0.3587	0.0348	0.0307	0.0452	0.0059	7	0.5931	0.0805	-0.0624	0.0674	0.0120
7	0.5130	0.0647	0.0258	0.0643	0.0109						
10	0.6761	0.1283	-0.0158	0.0864	0.0205						
15	0.7370	0.2191	-0.0677	0.1006	0.0333						
20	0.7370	0.3015	-0.0757	0.0996	0.0442						
25	0.7630	0.3902	-0.0819	0.1016	0.0545						
$M = 1.00$											
$M = 1.05$											
-10	-0.8224	0.1436	0.0233	-0.0927	0.0166	-7	-0.5612	0.0888	0.0681	-0.0615	0.0088
-7	-0.6585	0.0799	-0.0072	-0.0730	0.0092	-5	-0.4226	0.0564	0.0411	-0.0464	0.0055
-5	-0.4946	0.0399	-0.0139	-0.0531	0.0050	-3	-0.2817	0.0352	0.0253	-0.0300	0.0031
-3	-0.3248	0.0194	-0.0072	-0.0332	0.0025	-2	-0.1954	0.0277	0.0176	-0.0217	0.0022
-2	-0.2264	0.0139	-0.0010	-0.0222	0.0017	-1	-0.1045	0.0226	0.0080	-0.0097	0.0017
-1	-0.1281	0.0092	-0.0001	-0.0113	0.0011	0	-0.0068	0.0204	0.0001	0.0014	0.0013
0	-0.0387	0.0066	0.0076	-0.0014	0.0008	1	0.1045	0.0210	-0.0058	0.0138	0.0024
1	0.0685	0.0073	0.0109	0.0109	0.0011	2	0.1908	0.0252	-0.0124	0.0234	0.0029
2	0.1669	0.0107	0.0159	0.0215	0.0019	3	0.2817	0.0321	-0.0230	0.0336	0.0043
3	0.2562	0.0150	0.0244	0.0323	0.0029	5	0.4294	0.0511	-0.0399	0.0508	0.0073
5	0.4559	0.0337	0.0300	0.0531	0.0065	7	0.5680	0.0804	-0.0607	0.0664	0.0115
7	0.6227	0.0648	0.0206	0.0728	0.0113						
10	0.7419	0.1253	-0.0192	0.0904	0.0200						
15	0.7866	0.2242	-0.0768	0.0999	0.0335						
20	0.8164	0.3165	-0.0957	0.1013	0.0436						
$M = 0.90$											
$M = 1.10$											
-7	-0.7107	0.0993	0.0416	-0.0806	0.0114	-7	-0.5597	0.0887	0.0690	-0.0597	0.0086
-5	-0.5404	0.0529	0.0143	-0.0607	0.0059	-5	-0.4242	0.0556	0.0449	-0.0451	0.0052
-3	-0.3619	0.0239	-0.0018	-0.0394	0.0028	-3	-0.2842	0.0352	0.0276	-0.0290	0.0028
-2	-0.2518	0.0142	-0.0059	-0.0265	0.0017	-2	-0.1946	0.0285	0.0192	-0.0196	0.0020
-1	-0.1521	0.0097	0.0018	-0.0159	0.0010	-1	-0.1115	0.0237	0.0092	-0.0096	0.0015
0	-0.0420	0.0064	0.0087	-0.0040	0.0008	0	-0.0131	0.0207	0.0003	0.0005	0.0014
1	0.0787	0.0064	0.0148	0.0097	0.0011	1	0.0853	0.0223	-0.0078	0.0124	0.0019
2	0.1862	0.0097	0.0205	0.0213	0.0019	2	0.1749	0.0274	-0.0128	0.0224	0.0026
3	0.3121	0.0148	0.0144	0.0352	0.0038	3	0.2602	0.0342	-0.0236	0.0323	0.0037
5	0.5245	0.0413	-0.0126	0.0607	0.0089	5	0.4067	0.0546	-0.0414	0.0498	0.0070
7	0.6871	0.0767	-0.0404	0.0806	0.0143	7	0.5357	0.0828	-0.0601	0.0647	0.0108

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Continued

$$n^{ct} = 0.04 \quad \frac{c_f}{c} = 0.10$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.3849					-10	-0.7523	0.1640	0.0159	-0.0873	0.0178
-7	-0.2357					-7	-0.5745	0.0994	0.0028	-0.0632	0.0112
-5	-0.0930					-5	-0.3798	0.0610	-0.0334	-0.0416	0.0067
-3	0.0035					-3	-0.1972	0.0365	-0.0728	-0.0185	0.0042
-2	0.0695					-2	-0.0755	0.0299	-0.0885	-0.0043	0.0034
-1	0.1179					-1	0.0463	0.0293	-0.1091	0.0006	0.0034
0	0.1881					0	0.1753	0.0335	0.1206	0.0332	0.0042
1	0.2314					1	0.2946	0.0383	-0.1331	0.0369	0.0055
2	0.2920					2	0.4114	0.0467	-0.1395	-0.0236	0.0073
3	0.3503					3	0.5283	0.0575	-0.1475	0.0628	0.0096
5	0.4747										
7	0.6012										
10	0.7764										
15	0.8088										
20	0.7875										
25	0.7526										
$M = 0.60$											
-10	-0.4740	0.0826	-0.0525	-0.0567	0.0116	-10	-0.7444	0.1619	0.0460	-0.0850	0.0178
-7	-0.2953	0.0411	-0.0994	-0.0325	0.0059	-7	-0.5264	0.0969	0.0054	-0.0581	0.0106
-5	-0.1327	0.0231	-0.0890	-0.0130	0.0032	-5	-0.3734	0.0650	-0.0230	-0.0396	0.0075
-3	-0.0064	0.0216	-0.0863	0.0026	0.0026	-3	-0.2110	0.0450	-0.0474	-0.0206	0.0054
-2	0.0696	0.0221	-0.0894	0.0120	0.0025	-2	-0.1113	0.0394	-0.0598	-0.0088	0.0045
-1	0.1434	0.0274	-0.0795	0.0203	0.0030	-1	-0.0139	0.0365	-0.0714	0.0026	0.0043
0	0.2161	0.0321	-0.0789	0.0289	0.0036	0	0.0974	0.0376	-0.0865	0.0144	0.0045
1	0.2878	0.0353	-0.0792	0.0381	0.0046	1	0.2226	0.0410	-0.1033	0.0294	0.0056
2	0.3767	0.0426	-0.0673	0.0471	0.0058	2	0.3293	0.0468	-0.1192	0.0413	0.0066
3	0.4430	0.0532	-0.0651	0.0552	0.0065	3	0.4569	0.0582	-0.1399	0.0545	0.0085
5	0.5821	0.0853	-0.0510	0.0726	0.0104						
7	0.7041	0.1142	-0.0563	0.0906	0.0158						
10	0.8068	0.1784	-0.0951	0.1058	0.0245						
15	0.8325	0.1400	-0.1196	0.1133	0.0371						
20	0.8218	0.1858	-0.1184	0.1120	0.0493						
25	0.8218	0.2231	-0.1133	0.1104	0.0583						
$M = 0.80$											
-10	-0.5954	0.1096	-0.0757	-0.0712	0.0131	-7	-0.5034	0.0953	0.0065	-0.0558	0.0090
-7	-0.4048	0.0570	-0.0982	-0.0456	0.0067	-5	-0.3586	0.0652	-0.0214	-0.0385	0.0055
-5	-0.2305	0.0307	-0.0993	-0.0165	0.0040	-3	-0.2161	0.0487	-0.0422	-0.0211	0.0035
-3	-0.0660	0.0205	-0.0943	-0.0045	0.0027	-2	-0.1181	0.0422	-0.0514	-0.0101	0.0025
-2	0.0352	0.0205	-0.0921	0.0073	0.0025	-1	-0.0245	0.0405	-0.0614	0.0005	0.0023
-1	0.1305	0.0224	-0.0925	0.0182	0.0029	0	0.0757	0.0417	-0.0756	0.0127	0.0031
0	0.2258	0.0267	-0.0865	0.0296	0.0036	1	0.1827	0.0433	-0.0840	0.0245	0.0031
1	0.3461	0.0310	-0.0939	0.0438	0.0046	2	0.2762	0.0498	-0.1002	0.0355	0.0045
2	0.4356	0.0375	-0.0880	0.0534	0.0060	3	0.3920	0.0597	-0.1153	0.0476	0.0065
3	0.5309	0.0465	-0.0820	0.0641	0.0079	5	0.5769	0.0893	-0.1480	0.0693	0.0099
5	0.7025	0.0735	-0.0727	0.0845	0.0128						
$M = 0.90$											
-10	-0.7771	0.1466	-0.0107	-0.0842	0.0172	-7	-0.4845	0.0891	0.0079	-0.0538	0.0078
-7	-0.5318	0.0822	-0.0454	-0.0574	0.0093	-5	-0.3451	0.0617	-0.0177	-0.0374	0.0048
-5	-0.2486	0.0441	-0.0579	-0.0317	0.0049	-3	-0.2101	0.0448	-0.0370	-0.0207	0.0030
-3	-0.1239	0.0257	-0.1023	-0.0108	0.0031	-2	-0.1158	0.0401	-0.0442	-0.0106	0.0027
-2	-0.0026	0.0225	-0.1062	0.0033	0.0028	-1	-0.0236	0.0379	-0.0562	0.0000	0.0021
-1	0.1110	0.0241	-0.1112	0.0163	0.0030	0	0.0579	0.0432	-0.0631	0.0109	0.0024
0	0.2349	0.0283	-0.1123	0.0294	0.0038	1	0.1543	0.0437	-0.0755	0.0216	0.0033
1	0.3434	0.0321	-0.1109	0.0423	0.0049	2	0.2487	0.0511	-0.0904	0.0330	0.0044
2	0.4647	0.0403	-0.1172	0.0558	0.0068	3	0.3516	0.0611	-0.1046	0.0462	0.0058
3	0.5783	0.0523	-0.1215	0.0683	0.0091	5	0.5230	0.0880	-0.1333	0.0642	0.0094
$M = 1.05$											
-7	-0.5034	0.0953	0.0065	-0.0558	0.0090						
-5	-0.3586	0.0652	-0.0214	-0.0385	0.0055						
-3	-0.2161	0.0487	-0.0422	-0.0422	0.0035						
-2	-0.1181	0.0422	-0.0514	-0.0101	0.0025						
-1	-0.0245	0.0405	-0.0614	0.0005	0.0023						
0	0.0757	0.0417	-0.0756	0.0127	0.0031						
1	0.1827	0.0433	-0.0840	0.0245	0.0031						
2	0.2762	0.0498	-0.1002	0.0355	0.0045						
3	0.3920	0.0597	-0.1153	0.0476	0.0065						
5	0.5769	0.0893	-0.1480	0.0693	0.0099						
$M = 1.10$											
-7	-0.4845	0.0891	0.0079	-0.0538	0.0078						
-5	-0.3451	0.0617	-0.0177	-0.0374	0.0048						
-3	-0.2101	0.0448	-0.0370	-0.0207	0.0030						
-2	-0.1158	0.0401	-0.0442	-0.0106	0.0027						
-1	-0.0236	0.0379	-0.0562	0.0000	0.0021						
0	0.0579	0.0432	-0.0631	0.0109	0.0024						
1	0.1543	0.0437	-0.0755	0.0216	0.0033						
2	0.2487	0.0511	-0.0904	0.0330	0.0044						
3	0.3516	0.0611	-0.1046	0.0462	0.0058						
5	0.5230	0.0880	-0.1333	0.0642	0.0094						

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{c_f}{c} = 0.20$$

$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$	$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$
$M = 0.40$											
-10	-0.3288					-10	-0.6673	0.1482	-0.0273	-0.0769	0.0169
-7	-0.1270					-7	-0.4277	0.0835	-0.0657	-0.0486	0.0094
-5	0.0109					-5	-0.2273	0.0505	-0.1031	-0.0247	0.0056
-3	0.1302					-3	-0.0367	0.0354	-0.1377	-0.0019	0.0040
-2	0.1954					-2	0.1027	0.0324	-0.1625	0.0134	0.0042
-1	0.2529					-1	0.2273	0.0337	-0.1736	0.0271	0.0051
0	0.3169					0	0.3422	0.0571	-0.1754	0.0399	0.0064
1	0.3722					1	0.4522	0.0460	-0.1740	0.0519	0.0080
2	0.4287					2	0.5622	0.0559	-0.1768	0.0630	0.0102
3	0.4938					3	0.6599	0.0688	-0.1846	0.0732	0.0126
5	0.6186										
7	0.7271										
10	0.8139										
15	0.8356										
20	0.7987										
25	0.6728										
$M = 0.60$											
$M = 1.00$											
-10	-0.3898	0.0769	-0.0747	-0.0448	0.0086	-10	-0.6566	0.1477	-0.0032	-0.0744	0.0163
-7	-0.1611	0.0280	-0.1154	-0.0163	0.0028	-7	-0.4354	0.0876	-0.0438	-0.0495	0.0099
-5	0.0021	0.0125	-0.1148	0.0034	0.0014	-5	-0.2678	0.0590	-0.0756	-0.0288	0.0068
-3	0.1536	0.0137	-0.1141	0.0207	0.0016	-3	-0.0931	0.0430	-0.1069	-0.0080	0.0050
-2	0.2395	0.0169	-0.1108	0.0301	0.0024	-2	0.0116	0.0384	-0.1268	0.0044	0.0047
-1	0.3136	0.0193	-0.1068	0.0383	0.0032	-1	0.1281	0.0367	-0.1457	0.0177	0.0049
0	0.3823	0.0241	-0.1033	0.0468	0.0041	0	0.2678	0.0389	-0.1674	0.0327	0.0058
1	0.4511	0.0299	-0.0944	0.0541	0.0053	1	0.3958	0.0446	-0.1738	0.0463	0.0073
2	0.5262	0.0369	-0.0872	0.0624	0.0068	2	0.5053	0.0550	-0.1769	0.0579	0.0092
3	0.6014	0.0467	-0.0818	0.0709	0.0087	3	0.5984	0.0676	-0.1812	0.0680	0.0113
5	0.7346	0.0747	-0.0675	0.0886	0.0135						
7	0.8420	0.1125	-0.0727	0.1039	0.0195						
10	0.8914	0.1759	-0.1100	0.1140	0.0285						
15	0.8634	0.1363	-0.1223	0.1147	0.0417						
20	0.8656	0.1785	-0.1194	0.1082	0.0514						
25	0.8570	0.2208	-0.1153	0.1066	0.0617						
$M = 0.80$											
$M = 1.05$											
-10	-0.3974	0.0874	-0.0868	-0.0486	0.0098	-10	-0.6307	0.1408	-0.0044	-0.0712	0.0154
-7	-0.2178	0.0394	-0.1360	-0.0249	0.0041	-7	-0.4249	0.0881	-0.0404	-0.0475	0.0091
-5	-0.0324	0.0197	-0.1421	-0.0034	0.0023	-5	-0.2684	0.0594	-0.0679	-0.0288	0.0062
-3	0.1281	0.0169	-0.1368	0.0153	0.0023	-3	-0.1118	0.0440	-0.0921	-0.0105	0.0047
-2	0.2311	0.0177	-0.1361	0.0262	0.0031	-2	-0.0067	0.0401	-0.1092	0.0014	0.0045
-1	0.3165	0.0206	-0.1352	0.0355	0.0040	-1	0.0917	0.0390	-0.1229	0.0131	0.0046
0	0.4062	0.0249	-0.1292	0.0456	0.0051	0	0.2125	0.0401	-0.1446	0.0266	0.0055
1	0.4931	0.0313	-0.1205	0.0547	0.0065	1	0.3511	0.0440	-0.1640	0.0424	0.0067
2	0.5785	0.0376	-0.1094	0.0652	0.0083	2	0.4652	0.0539	-0.1717	0.0538	0.0085
3	0.6727	0.0481	-0.1006	0.0759	0.0106	3	0.5591	0.0660	-0.1776	0.0641	0.0105
5	0.8596	0.0798	-0.1002	0.0965	0.0164						
$M = 0.90$											
$M = 1.10$											
-10	-0.5506	0.0854	-0.0814	-0.0678	0.0137	-10	-0.6091	0.1371	0.0024	-0.0686	0.0143
-7	-0.3433	0.0656	-0.1084	-0.0403	0.0070	-7	-0.4089	0.0836	-0.0389	-0.0462	0.0086
-5	-0.1684	0.0376	-0.1278	-0.0177	0.0040	-5	-0.2626	0.0588	-0.0630	-0.0291	0.0060
-3	0.0324	0.0249	-0.1461	0.0059	0.0031	-3	-0.1141	0.0439	-0.0858	-0.0111	0.0045
-2	0.1581	0.0236	-0.1525	0.0187	0.0034	-2	-0.0215	0.0386	-0.1003	0.0000	0.0043
-1	0.2695	0.0264	-0.1554	0.0309	0.0044	-1	0.2949	0.0402	-0.1687	0.0111	0.0045
0	0.3692	0.0306	-0.1482	0.0423	0.0057	0	0.1786	0.0429	-0.1280	0.0225	0.0052
1	0.4677	0.0376	-0.1479	0.0535	0.0073	1	0.2992	0.0466	-0.1464	0.0359	0.0064
2	0.5830	0.0472	-0.1482	0.0668	0.0097	2	0.3982	0.0545	-0.1553	0.0473	0.0082
3	0.6996	0.0605	-0.1525	0.0786	0.0122	3	0.5187	0.0667	-0.1697	0.0607	0.0100

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Continued

$$\frac{t}{c} = 0.04 \quad \frac{C_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.95$											
-10	-0.2995					-7	-0.3324	0.0423	0.0597	-0.0346	0.0100
-7	-0.0803					-5	-0.1247	0.0290	0.0097	-0.0097	0.0068
-5	0.0499					-3	0.0733	0.0233	0.0381	0.0134	0.0058
-3	0.1693					-2	0.2114	0.0236	0.0701	0.0278	0.0061
-2	0.2344					-1	0.3178	0.0255	0.0941	0.0392	0.0071
-1	0.2952					0	0.4253	0.0293	0.1177	0.0519	0.0084
0	0.3646					1	0.5231	0.0332	0.1386	0.0630	0.0104
1	0.4232					2	0.6355	0.0394	0.1637	0.0736	-0.0127
2	0.4797										
3	0.5426										
5	0.6685										
7	0.7640										
10	0.8400										
15	0.8378										
20	0.7944										
25	0.7683										
$M = 1.00$											
-10	-0.3351	0.0708	-0.0700	-0.0379	0.0085	-7	-0.3459	0.0873	-0.0828	-0.0376	0.0101
-7	-0.1010	0.0293	-0.1178	-0.0089	0.0032	-5	-0.1735	0.0636	-0.1164	-0.0159	0.0075
-5	0.0677	0.0190	-0.1162	0.0108	0.0022	-3	-0.0047	0.0524	-0.1463	0.0041	0.0063
-3	0.2073	0.0224	-0.1168	0.0273	0.0026	-2	0.1165	0.0501	-0.1672	0.0182	0.0062
-2	0.3007	0.0259	-0.1111	0.0370	0.0032	-1	0.2492	0.0510	-0.1877	0.0323	0.0066
-1	0.3737	0.0317	-0.1111	0.0456	0.0040	0	0.3913	0.0558	-0.1912	0.0477	0.0080
0	0.4489	0.0383	-0.1051	0.0546	0.0050	1	0.4868	0.0653	-0.1947	0.0580	0.0091
1	0.5133	0.0449	-0.1047	0.0622	0.0063	2	0.5893	0.0773	-0.1962	0.0684	0.0113
2	0.5928	0.0528	-0.0951	0.0701	0.0080						
3	0.6551	0.0639	-0.0931	0.0782	0.0099						
5	0.7883	0.0948	-0.0767	0.0961	0.0148						
7	0.8871	0.1321	-0.0845	0.1095	0.0209						
10	0.9128	0.2018	-0.1207	0.1173	0.0262						
15	0.8871	0.2936	-0.1441	0.1180	0.0451						
20	0.8720	0.3834	-0.1456	0.1140	0.0539						
25	0.8828	0.4711	-0.1518	0.1124	0.0621						
$M = 0.80$											
$M = 1.05$											
-10	-0.3378	0.0807	-0.0843	-0.0391	0.0097	-7	-0.3377	0.0841	-0.0787	-0.0366	0.0090
-7	-0.1317	0.0369	-0.1433	-0.0123	0.0041	-5	-0.1789	0.0632	-0.1086	-0.0174	0.0065
-5	0.0611	0.0232	-0.1449	0.0095	0.0028	-3	-0.0268	0.0522	-0.1330	0.0013	0.0059
-3	0.2156	0.0246	-0.1423	0.0156	0.0035	-2	0.0816	0.0517	-0.1497	0.0133	0.0057
-2	0.3076	0.0283	-0.1397	0.0374	0.0043	-1	0.1990	0.0522	-0.1695	0.0267	0.0059
-1	0.3915	0.0315	-0.1364	0.0463	0.0052	0	0.3388	0.0564	-0.1867	0.0416	0.0067
0	0.4607	0.0380	-0.1050	0.0550	0.0066	1	0.4450	0.0652	-0.1907	0.0543	0.0079
1	0.5446	0.0453	-0.1152	0.0650	0.0083	2	0.5323	0.0756	-0.1893	0.0551	0.0096
2	0.6285	0.0550	-0.1148	0.0748	0.0100	3	0.6284	0.1045	-0.1694	0.0729	0.0119
3	0.7448	0.0687	-0.1185	0.0860	0.0130						
5	0.9067	0.1032	-0.1181	0.1061	0.0186						
$M = 0.90$											
$M = 1.10$											
-7	-0.2359	0.0663	-0.1473	-0.0246	0.0076	-7	-0.3337	0.0807	-0.0725	-0.0363	0.0081
-5	-0.0415	0.0446	-0.1693	-0.0010	0.0051	-5	-0.1873	0.0614	-0.0997	-0.0188	0.0058
-3	0.1594	0.0389	-0.1826	0.0226	0.0049	-3	-0.0312	0.0519	-0.1237	-0.0004	0.0049
-2	0.2761	0.0427	-0.1847	0.0344	0.0057	-2	0.0624	0.0511	-0.1384	0.0110	0.0049
-1	0.3797	0.0462	-0.1843	0.0462	0.0069	-1	0.1593	0.0524	-0.1527	0.0216	0.0055
0	0.4705	0.0542	-0.1835	0.0570	0.0084	0	0.2799	0.0569	-0.1712	0.0351	0.0063
1	0.5651	0.0631	-0.1809	0.0678	0.0104	1	0.4015	0.0640	-0.1840	0.0498	0.0074
2	0.6714	0.0711	-0.1719	0.0777	0.0122	2	0.4930	0.0741	-0.1863	0.0596	0.0090
3	0.7465	0.0844	-0.1731	0.0865	0.0146	3	0.5877	0.0900	-0.1944	0.0694	0.0106

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Continued

$$\frac{c_f}{c} = 0.04 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$	$\alpha$ , deg	$c_L$	$c_D$	$c_M$	$c_l$	$c_n$
$M = 0.40$											
$M = 0.55$											
-10	-1875					-10	-4715	.1297	=.0965	=.0521	.0155
-7	-0066					-7	-2544	.0854	=.1422	=.0243	.0099
-5	-1300					-5	-0310	.0635	=.1748	.0011	.0073
-3	-2423					-3	-1799	.0565	=.2027	.0250	.0068
-2	-3084					-2	-3102	.0579	=.2060	.0392	.0075
-1	-3657					-1	-4343	.0641	=.2064	.0520	.0088
0	-4317					0	-5211	.0735	=.2010	.0623	.0104
1	-4846					1	-6179	.0839	=.2049	.0727	.0123
2	-5595					2	-7320	.0976	=.2099	.0823	.0147
3	-6080										
5	-7379										
7	-8018										
10	-8415										
15	-8150										
20	-7930										
25	-7820										
$M = 0.60$											
$M = 1.00$											
-10	-2618	.0577	-0.727	-0.0301	.0075	-10	-4880	.1307	=.0669	=.0545	.0145
-7	-0145	.0223	-0.1128	.0005	.0028	-7	-2706	.0868	=.1152	=.0276	.0098
-5	-1527	.0201	-0.1079	.0204	.0024	-5	-0969	.0677	=.1462	=.0061	.0076
-3	-3000	.0241	-0.1072	.0373	.0034	-3	-0945	.0601	=.1769	.0163	.0067
-2	-3928	.0300	-0.0987	.0473	.0044	-2	-2481	.0590	=.1954	.0330	.0070
-1	-4430	.0357	-0.0963	.0543	.0052	-1	-3722	.0639	=.2005	.0462	.0077
0	-5150	.0443	-0.0910	.0627	.0065	0	-4667	.0727	=.2003	.0572	.0090
1	-5804	.0524	-0.0913	.0708	.0082	1	-5636	.0828	=.1994	.0679	.0110
2	-6481	.0625	-0.0819	.0798	.0101	2	-6617	.0965	=.2020	.0769	.0132
3	-7266	.0759	-0.0756	.0887	.0125						
5	-8510	.1094	-0.0683	.1053	.0180						
7	-9099	.1556	-0.0814	.1155	.0245						
10	-9055	.2227	-0.1228	.1195	.0333						
15	-8750	.3187	-0.1402	.1155	.0452						
20	-8641	.4078	-0.1414	.1142	.0547						
25	-8641	.5011	-0.1449	.1139	.0658						
$M = 0.80$											
$M = 1.05$											
-10	-2728	.0717	-0.0839	-0.0314	.0087	-10	-4732	.1264	=.0638	=.0532	.0132
-7	-0314	.0344	-0.1475	-0.0016	.0039	-7	-2780	.0862	=.1123	=.0288	.0086
-5	-1532	.0300	-0.1482	.0200	.0033	-5	-1123	.0684	=.1356	=.0088	.0066
-3	-3139	.0342	-0.1453	.0381	.0048	-3	-0624	.0611	=.1616	.0119	.0060
-2	-3961	.0377	-0.1379	.0477	.0057	-2	-1872	.0608	=.1800	.0260	.0062
-1	-4634	.0448	-0.1375	.0569	.0070	-1	-3155	.0636	=.1946	.0406	.0066
0	-5456	.0493	-0.1335	.0664	.0084	0	-4256	.0726	=.1993	.0532	.0078
1	-6203	.0616	-0.1235	.0762	.0105	1	-5163	.0832	=.1953	.0634	.0093
2	-7175	.0726	-0.1158	.0864	.0128	2	-6128	.0957	=.1997	.0725	.0112
3	-4081	.0854	-0.0715	.0959	.0154						
$M = 0.90$											
$M = 1.10$											
-10	-3355	.1045	-0.1479	-0.0368	.0122	-10	-4479	.1168	=.0677	=.0512	.0125
-7	-1066	.0663	-0.1797	-0.0089	.0075	-7	-2655	.0806	=.1014	=.0283	.0079
-5	-0921	.0518	-0.2023	.0141	.0057	-5	-1092	.0664	=.1303	=.0090	.0060
-3	-2829	.0501	-0.2067	.0352	.0065	-3	-0546	.0591	=.1536	.0101	.0056
-2	-3842	.0557	-0.1981	.0465	.0077	-2	-1584	.0599	=.1666	.0224	.0061
-1	-4684	.0631	-0.1976	.0573	.0092	-1	-2677	.0645	=.1802	.0350	.0066
0	-5592	.0712	-0.1929	.0681	.0108	0	-3824	.0714	=.1880	.0486	.0075
1	-6579	.0799	-0.1873	.0780	.0129	1	-4807	.0819	=.1906	.0598	.0091
2	-7553	.0899	-0.1808	.0870	.0148	2	-5856	.0959	=.1912	.0698	.0107

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Continued

 $\frac{k}{c} = 0.06$        $\frac{c_f}{c} = \text{NONE}$ 

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$						$M = 0.95$					
-10	-0.5810					-10	-0.8571	0.1739	0.1523	-0.1044	-0.0258
-7	-0.4900					-7	-0.6015	0.0958	0.0765	-0.0726	-0.0151
-5	-0.3240					-5	-0.4426	0.0628	0.0438	-0.0537	-0.0088
-3	-0.2008					-3	-0.2805	0.0421	0.0266	-0.0340	-0.0052
-2	-0.1339					-2	-0.1714	0.0299	0.0041	-0.0208	-0.0034
-1	-0.0803					-1	-0.0841	0.0284	0.0041	-0.0102	-0.0018
0	-0.0134					0	-0.0000	0.0284	0.0024	0.0000	-0.0015
1	0.0536					1	0.0873	0.0284	-0.0027	0.0106	-0.0015
2	0.1125					2	0.1714	0.0337	-0.0079	0.0204	-0.0019
3	0.1767					3	0.2649	0.0452	-0.0183	0.0318	-0.0027
5	0.2972					5	0.4207	0.0667	-0.0338	0.0507	-0.0046
7	0.4338					7	0.5922	0.1088	-0.0717	0.0711	-0.0088
10	0.6159					10	0.8633	0.1917	-0.1475	0.1029	-0.0187
15	0.7230					15	1.1376	0.3425	-0.1985	0.1392	-0.0379
20	0.7256					20	1.1064	0.4399	-0.2054	0.1324	-0.0509
25	0.7149					25	1.1064	0.5702	-0.2399	0.1343	-0.0663
$M = 0.60$						$M = 1.00$					
-10	-0.6492	0.1276	0.0447	-0.0823	-0.0172	-10	-0.7676	0.1660	0.1316	-0.0921	-0.0238
-7	-0.5002	0.0585	-0.0079	-0.0602	-0.0083	-7	-0.5397	0.1003	0.0769	-0.0648	-0.0142
-5	-0.3635	0.0282	-0.0048	-0.0428	-0.0041	-5	-0.3988	0.0685	0.0477	-0.0473	-0.0099
-3	-0.2214	0.0161	-0.0048	-0.0254	-0.0021	-3	-0.2519	0.0487	0.0249	-0.0306	-0.0064
-2	-0.1449	0.0115	-0.0060	-0.0166	-0.0012	-2	-0.1709	0.0435	0.0146	-0.0211	-0.0047
-1	-0.0724	0.0101	-0.0012	-0.0083	0.0000	-1	-0.0930	0.0391	0.0080	-0.0120	-0.0030
0	0.0000	0.0074	-0.0043	0.0003	0.0000	0	-0.0030	0.0384	0.0040	-0.0015	-0.0025
1	0.0752	0.0082	0.0012	0.0095	0.0002	1	0.0840	0.0399	-0.0023	0.0084	-0.0030
2	0.1435	0.0107	0.0000	0.0168	0.0003	2	0.1649	0.0442	-0.0100	0.0186	-0.0030
3	0.2187	0.0148	-0.0012	0.0257	0.0003	3	0.2519	0.0523	-0.0215	0.0288	-0.0035
5	0.3567	0.0262	0.0024	0.0423	-0.0006	5	0.3958	0.0723	-0.0421	0.0448	-0.0051
7	0.5057	0.0552	0.0012	0.0599	-0.0018	7	0.5487	0.1069	-0.0766	0.0630	-0.0086
10	0.6970	0.1225	-0.0290	0.0839	-0.0088	10	0.7886	0.1748	-0.1307	0.0913	-0.0164
15	0.7790	0.2318	-0.1004	0.1002	-0.0230	15	1.1124	0.3310	-0.2142	0.1332	-0.0356
20	0.7653	0.2520	-0.1288	0.0979	-0.0347	20	1.3014	0.5087	-0.2374	0.1587	-0.0589
25	0.7817	0.4141	-0.1305	0.0975	-0.0464	25	1.2234	0.6105	-0.2739	0.1514	-0.0729
$M = 0.80$						$M = 1.05$					
-10	-0.7464	0.1249	0.0253	-0.0931	-0.0186	-10	-0.7513	0.1634	0.1275	-0.0905	-0.0241
-7	-0.5820	0.0623	0.0029	-0.0702	-0.0108	-7	-0.5519	0.0995	0.0812	-0.0663	-0.0151
-5	-0.4379	0.0327	-0.0135	-0.0511	-0.0054	-5	-0.4103	0.0696	0.0543	-0.0491	-0.0108
-3	-0.2568	0.0155	-0.0163	-0.0296	-0.0026	-3	-0.2572	0.0497	0.0336	-0.0316	-0.0066
-2	-0.1755	0.0122	-0.0094	-0.0200	-0.0014	-2	-0.1734	0.0455	0.0236	-0.0214	-0.0052
-1	-0.0794	0.0105	-0.0041	-0.0090	-0.0004	-1	-0.0809	0.0426	0.0144	-0.0130	-0.0039
0	0.0000	0.0091	-0.0033	0.0005	0.0003	0	-0.0058	0.0412	0.0038	-0.0021	-0.0034
1	0.0868	0.0109	-0.0024	0.0101	0.0001	1	0.0867	0.0419	-0.0083	0.0084	-0.0037
2	0.1718	0.0140	0.0021	0.0197	-0.0001	2	0.1734	0.0470	-0.0186	0.0182	-0.0038
3	0.2531	0.0190	0.0090	0.0291	-0.0002	3	0.2572	0.0548	-0.0291	0.0281	-0.0041
5	0.4249	0.0373	0.0123	0.0489	-0.0015	5	0.4103	0.0746	-0.0495	0.0456	-0.0060
7	0.5820	0.0722	-0.0024	0.0695	-0.0042	7	0.5548	0.1086	-0.0754	0.0635	-0.0096
10	0.7484	0.1339	-0.0257	0.0919	-0.0116	10	0.7542	0.1719	-0.1247	0.0880	-0.0171
15	0.7945	0.2444	-0.1075	0.1005	-0.0518	15	1.0836	0.3304	-0.2116	0.1287	-0.0364
20	0.8129	0.3379	-0.1414	0.1009	-0.0385	20	1.3206	0.5343	-0.2749	0.1589	-0.0612
25	0.8406	0.4397	-0.1504	0.1031	-0.0511	25	1.2425	0.6295	-0.2845	0.1620	-0.0791
$M = 0.90$						$M = 1.10$					
-10	-0.7844	0.1461	0.0824	-0.0996	-0.0225	-10	-0.7282	0.1566	0.1277	-0.0897	-0.0245
-7	-0.6072	0.0827	0.0515	-0.0757	-0.0135	-7	-0.5399	0.0940	0.0818	-0.0662	-0.0156
-5	-0.4595	0.0504	0.0261	-0.0558	-0.0072	-5	-0.4070	0.0647	0.0588	-0.0507	-0.0114
-3	-0.2855	0.0263	0.0116	-0.0351	-0.0032	-3	-0.2492	0.0476	0.0368	-0.0319	-0.0073
-2	-0.1969	0.0190	0.0043	-0.0231	-0.0022	-2	-0.1661	0.0422	0.0258	-0.0225	-0.0060
-1	-0.0985	0.0169	0.0026	-0.0120	-0.0012	-1	-0.0831	0.0408	0.0144	-0.0131	-0.0044
0	0.0098	0.0138	-0.0011	0.008	0.0006	0	0.0000	0.0401	0.0022	-0.0034	-0.0041
1	0.0985	0.0158	-0.0047	0.0120	-0.0008	1	0.0914	0.0408	-0.0086	0.0067	-0.0044
2	0.1969	0.0210	-0.0083	0.0231	-0.0007	2	0.1661	0.0450	-0.0193	0.0161	-0.0043
3	0.2954	0.0290	-0.0145	0.0351	-0.0010	3	0.2575	0.0525	-0.0300	0.0252	-0.0047
5	0.4529	0.0525	-0.0283	0.0546	-0.0031	5	0.4070	0.0728	-0.0530	0.0427	-0.0064
7	0.6138	0.0916	-0.0519	0.0749	-0.0073	7	0.5538	0.1055	-0.0799	0.0605	-0.0105
10	0.7976	0.1574	-0.0845	0.0992	-0.0156	10	0.7421	0.1668	-0.1203	0.0837	-0.0174
15	0.9682	0.2801	-0.1245	0.1215	-0.0312	15	1.0522	0.3180	-0.2058	0.1216	-0.0355
20	0.9288	0.3786	-0.1735	0.1139	-0.0431	20	1.2820	0.5161	-0.2714	0.1512	-0.0589
25	0.9846	0.4956	-0.1910	0.1155	-0.0569						

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.20$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.2633					-10	-0.6336	0.1469	0.0135	-0.0763	-0.0205
-7	-0.0967					-7	-0.4396	0.0900	-0.0235	-0.0516	-0.0126
-5	0.0242					-5	-0.2941	0.0662	-0.0460	-0.0336	-0.0076
-3	0.1666					-3	-0.1189	0.0462	-0.0737	-0.122	-0.0045
-2	0.2257					-2	0.0110	0.0438	-0.1014	0.0025	-0.0036
-1	0.2982					-1	0.1549	0.0454	-0.1377	0.0186	-0.0035
0	0.3519					0	0.2879	0.0538	-0.1654	0.0338	-0.0045
1	0.4137					1	0.4177	0.0630	-0.1876	0.0490	-0.0060
2	0.4594					2	0.5225	0.0385	-0.1751	0.0613	-0.0077
3	0.5215					3	0.6242	0.0939	-0.2167	0.0737	-0.0098
5	0.6340					5	0.7776	0.1284	-0.2339	0.0923	-0.0145
7	0.7415					7	0.9356	0.2570	-0.2561	0.1097	-0.0208
10	0.8650					10	1.1171	0.2671	-0.2893	0.1321	-0.0319
15	0.8919					15	1.3486	0.4309	-0.3101	0.1606	-0.0544
20	0.8465					20	1.2015	0.5110	-0.2963	0.1416	-0.0644
25	0.8194					25	1.1390	0.6109	-0.2893	0.1341	-0.0771
$M = 0.60$											
-10	-0.3235	0.0790	-0.0906	-0.0389	-0.0096	-10	-0.6129	0.1441	0.0213	-0.0727	-0.0200
-7	-0.1110	0.0236	-0.1280	-0.0115	-0.0020	-7	-0.4146	0.0939	-0.0259	-0.0469	-0.0130
-5	0.0234	0.0167	-0.1292	-0.0003	-0.0032	-5	-0.2569	0.0709	-0.0541	-0.0288	-0.0085
-3	0.1590	0.0181	-0.1261	0.0196	-0.0007	-3	-0.0916	0.0584	-0.0834	-0.0089	-0.0059
-2	0.2207	0.0203	-0.1231	0.0263	-0.0006	-2	0.0000	0.0539	-0.0990	0.0015	-0.0052
-1	0.2933	0.0236	-0.1219	0.0346	-0.0009	-1	0.1067	0.0547	-0.1173	0.0135	-0.0052
0	0.3646	0.0296	-0.1213	0.0432	-0.0012	0	0.2238	0.0569	-0.1389	0.0261	-0.0052
1	0.4674	0.0370	-0.1280	0.0562	-0.0018	1	0.3545	0.0643	-0.1648	0.0407	-0.0060
2	0.5236	0.0430	-0.1189	0.0625	-0.0027	2	0.4567	0.0754	-0.1868	0.0534	-0.0076
3	0.5880	0.0518	-0.1170	0.0700	-0.0035	3	0.5618	0.0916	-0.2047	0.0662	-0.0092
5	0.7320	0.0768	-0.1098	0.0818	-0.0067	5	0.7301	0.1256	-0.2313	0.0853	-0.0135
7	0.8526	0.1179	-0.1122	0.1011	-0.0115	7	0.8893	0.1752	-0.2519	0.1032	-0.0197
10	0.9239	0.1900	-0.1503	0.1131	-0.0210	10	1.0636	0.2564	-0.2811	0.1243	-0.0301
15	0.9431	0.3013	-0.1898	0.1201	-0.0359	15	1.3040	0.4241	-0.3210	0.1542	-0.0261
20	0.9019	0.1955	-0.2007	0.1115	-0.0488	20	1.4241	0.5940	-0.3363	0.1725	-0.0375
25	0.8965	0.2434	-0.2086	0.1091	-0.0604						
$M = 0.80$											
-10	-0.4393	0.0875	-0.0906	-0.0528	-0.0116	-10	-0.5851	0.1389	0.0176	-0.0691	-0.0193
-7	-0.2465	0.0415	-0.1209	-0.0283	-0.0094	-7	-0.3953	0.0805	-0.0250	-0.0455	-0.0129
-5	-0.0760	0.0247	-0.1316	-0.0087	-0.0022	-5	-0.2491	0.0691	-0.0506	-0.0283	-0.0085
-3	0.1094	0.0206	-0.1332	0.0131	-0.0010	-3	-0.0912	0.0584	-0.0779	-0.0097	-0.0062
-2	0.1983	0.0228	-0.1304	0.0232	-0.0010	-2	0.0058	0.0549	-0.0938	0.0012	-0.0055
-1	0.2855	0.0282	-0.1341	0.0335	-0.0013	-1	0.1028	0.0556	-0.1099	0.0128	-0.0052
0	0.3785	0.0337	-0.1353	0.0439	-0.0021	0	0.2013	0.0591	-0.1275	0.0239	-0.0054
1	0.4671	0.0419	-0.1398	0.0547	-0.0027	1	0.3027	0.0669	-0.1467	0.0357	-0.0062
2	0.5728	0.0510	-0.1407	0.0677	-0.0040	2	0.4011	0.0762	-0.1653	0.0471	-0.0075
3	0.6655	0.0625	-0.1365	0.0783	-0.0053	3	0.5127	0.0898	-0.1845	0.0596	-0.0093
5	0.8008	0.0921	-0.1365	0.0952	-0.0089	5	0.6821	0.1240	-0.2210	0.0801	-0.0136
7	0.9157	0.1368	-0.1427	0.1116	-0.0146	7	0.8428	0.1689	-0.2441	0.0988	-0.0194
10	1.0529	0.1172	-0.1530	0.1291	-0.0239	10	1.0195	0.2472	-0.2678	0.1195	-0.0300
15	0.9750	0.3146	-0.2115	0.1192	-0.0387	15	1.2657	0.4188	-0.3216	0.1497	-0.0512
20	0.9605	0.4139	-0.2190	0.1152	-0.0457	20	1.4308	0.6124	-0.3601	0.1722	-0.0766
25	0.9602	0.5196	-0.2305	0.1138	-0.0645						
$M = 0.90$											
-10	-0.5699	0.1244	-0.0248	-0.0722	-0.0244	-10	-0.5632	0.1351	0.0206	-0.0658	-0.0191
-7	-0.3936	0.0697	-0.0557	-0.0464	-0.0091	-7	-0.3870	0.0886	-0.0199	-0.0436	-0.0124
-5	-0.2273	0.0445	-0.0867	-0.0254	-0.0052	-5	-0.2469	0.0683	-0.0454	-0.0274	-0.0090
-3	0.0000	0.0308	-0.1286	0.0006	-0.0024	-3	-0.0916	0.0559	-0.0746	-0.0088	-0.0065
-2	0.1301	0.0311	-0.1469	0.0156	-0.0023	-2	0.0042	0.0526	-0.0899	0.0013	-0.0058
-1	0.2372	0.0352	-0.1523	0.0278	-0.0024	-1	0.0943	0.0540	-0.1053	0.0114	-0.0056
0	0.3278	0.0413	-0.1534	0.0390	-0.0031	0	0.1859	0.0259	-0.1218	0.0219	-0.0056
1	0.4348	0.0511	-0.1577	0.0514	-0.0043	1	0.2816	0.0627	-0.1374	0.0327	-0.0063
2	0.5287	0.0616	-0.1650	0.0622	-0.0054	2	0.3787	0.0730	-0.1553	0.0439	-0.0076
3	0.6325	0.0777	-0.1741	0.0746	-0.0074	3	0.4689	0.0853	-0.1731	0.0547	-0.0090
5	0.7791	0.1110	-0.1945	0.0932	-0.0120	5	0.6381	0.1160	-0.2056	0.0744	-0.0131
7	0.9289	0.1608	-0.2098	0.1111	-0.0187	7	0.8018	0.1618	-0.2320	0.0929	-0.0190
10	1.0805	0.2365	-0.2244	0.1307	-0.0291	10	0.9683	0.2354	-0.2596	0.1121	-0.0285
15	1.1002	0.3500	-0.2426	0.1331	-0.0440	15	1.2041	0.3984	-0.3081	0.1421	-0.0490
20	1.0574	0.4488	-0.2500	0.1251	-0.0565						
25	1.0343	0.5574	-0.2594	0.1231	-0.0699						
$M = 1.10$											
-10	-0.5699	0.1244	-0.0248	-0.0722	-0.0244	-10	-0.5632	0.1351	0.0206	-0.0658	-0.0191
-7	-0.3936	0.0697	-0.0557	-0.0464	-0.0091	-7	-0.3870	0.0886	-0.0199	-0.0436	-0.0124
-5	-0.2273	0.0445	-0.0867	-0.0254	-0.0052	-5	-0.2469	0.0683	-0.0454	-0.0274	-0.0090
-3	0.0000	0.0308	-0.1286	0.0006	-0.0024	-3	-0.0916	0.0559	-0.0746	-0.0088	-0.0065
-2	0.1301	0.0311	-0.1469	0.0156	-0.0023	-2	0.0042	0.0526	-0.0899	0.0013	-0.0058
-1	0.2372	0.0352	-0.1523	0.0278	-0.0024	-1	0.0943	0.0540	-0.1053	0.0114	-0.0056
0	0.3278	0.0413	-0.1534	0.0390	-0.0031	0	0.1859	0.0259	-0.1218	0.0219	-0.0056
1	0.4348	0.0511	-0.1577	0.0514	-0.0043	1	0.2816	0.0627	-0.1374	0.0327	-0.0063
2	0.5287	0.0616	-0.1650	0.0622	-0.0054	2	0.3787	0.0730	-0.1553	0.0439	-0.0076
3	0.6325	0.0777	-0.1741	0.0746	-0.0074	3	0.4689	0.0853	-0.1731	0.0547	-0.0090
5	0.7791	0.1110	-0.1945	0.0932	-0.0120	5	0.6381	0.1160	-0.2056	0.0744	-0.0131
7	0.9289	0.1608	-0.2098	0.1111	-0.0187	7	0.8018	0.1618	-0.2320	0.0929	-0.0190
10	1.0805	0.2365	-0.2244	0.1307	-0.0291	10	0.9683	0.2354	-0.2596	0.1121	-0.0285
15	1.1002	0.3500	-0.2426	0.1331	-0.0440	15	1.2041	0.3984	-0.3081	0.1421	-0.0490
20	1.0574	0.4488	-0.2500	0.1251	-0.0565						
25	1.0343	0.5574	-0.2594	0.1231	-0.0699						

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.55$											
-10	-0.2280					-10	-0.5561	0.1403	-0.0377	-0.0654	-0.0192
-7	-0.0510					-7	-0.3530	0.0911	-0.0757	-0.0389	-0.0116
-5	0.1019					-5	-0.1968	0.0692	-0.0982	-0.0203	-0.0076
-3	0.2360					-3	0.0359	0.0550	-0.1448	0.0047	-0.0050
-2	0.2816					-2	0.1718	0.0576	-0.1741	0.0224	-0.0050
-1	0.3406					-1	0.3046	0.0669	-0.2066	0.0379	-0.0055
0	0.4103					0	0.4280	0.0750	-0.2163	0.0521	-0.0070
1	0.4613					1	0.5264	0.0884	-0.2260	0.0711	-0.0085
2	0.5176					2	0.6217	0.1018	-0.2308	0.0745	-0.0103
3	0.5686					3	0.7107	0.1190	-0.2412	0.0853	-0.0128
5	0.6839					5	0.8497	0.1567	-0.2550	0.1024	-0.0171
7	0.8046					7	0.9934	0.2127	-0.2723	0.1202	-0.0238
10	0.8958					10	1.1558	0.2957	-0.2826	0.1395	-0.0348
15	0.8931					15	1.3433	0.4533	-0.2965	0.1649	-0.0554
20	0.8556					20	1.0215	0.5223	-0.2792	0.1422	-0.0641
25	0.8234					25	1.0777	0.5946	-0.2619	0.1346	-0.0763
$M = 0.60$											
$M = 1.00$											
-10	-0.2669	0.0775	-0.0973	-0.0332	-0.0074	-10	-0.5474	0.1390	-0.0252	-0.0641	-0.0183
-7	-0.0520	0.0268	-0.126	-0.0050	-0.0014	-7	-0.3449	0.0940	-0.0683	-0.0382	-0.0115
-5	0.0986	0.0222	-0.1296	-0.0131	-0.0006	-5	-0.1920	0.0741	-0.0975	-0.0191	-0.0081
-3	0.2532	0.0249	-0.1338	0.0309	-0.0006	-3	-0.0075	0.0645	-0.1290	0.0027	-0.0065
-2	0.3409	0.0277	-0.1278	0.0399	-0.0006	-2	0.0975	0.0645	-0.1483	0.0151	-0.0059
-1	0.4011	0.0337	-0.1326	0.0478	-0.0009	-1	0.2235	0.0678	-0.1719	0.0273	-0.0059
0	0.4750	0.0397	-0.1326	0.0565	-0.0014	0	0.3999	0.0748	-0.2017	0.0437	-0.0067
1	0.5311	0.0452	-0.1254	0.0638	-0.0024	1	0.4724	0.0859	-0.2176	0.0573	-0.0085
2	0.5914	0.0517	-0.1193	0.0706	-0.0033	2	0.5624	0.0991	-0.2249	0.0683	-0.0102
3	0.6571	0.0641	-0.1144	0.0791	-0.0047	3	0.6524	0.1162	-0.2342	0.0790	-0.0122
5	0.8022	0.0909	-0.1115	0.0944	-0.0077	5	0.8069	0.1530	-0.2481	0.0965	-0.0171
7	0.9117	0.1339	-0.1115	0.1090	-0.0091	7	0.9568	0.2080	-0.2674	0.1147	-0.0235
10	0.9500	0.2086	-0.1508	0.1173	-0.0233	10	1.0218	0.2942	-0.2926	0.1351	-0.0344
15	0.9309	0.3118	-0.1872	0.1180	-0.0366	15	1.3498	0.4646	-0.3244	0.1638	-0.0570
20	0.9035	0.4079	-0.1932	0.1123	-0.0490						
25	0.8980	0.5021	-0.1992	0.1100	-0.0606						
$M = 0.80$											
$M = 1.05$											
-10	-0.3339	0.0820	-0.1223	-0.0387	-0.0097	-10	-0.5349	0.1365	-0.0230	-0.0614	-0.0187
-7	-0.1471	0.0409	-0.1493	-0.0146	-0.0043	-7	-0.3383	0.0921	-0.0669	-0.0368	-0.0117
-5	0.0296	0.0259	-0.1530	0.0056	-0.0017	-5	-0.1879	0.0729	-0.0892	-0.0193	-0.0087
-3	0.2007	0.0305	-0.1490	0.0261	-0.0017	-3	-0.0101	0.0622	-0.1218	0.0018	-0.0065
-2	0.2821	0.0340	-0.1465	0.0344	-0.0020	-2	0.0896	0.0619	-0.1388	0.0132	-0.0061
-1	0.3691	0.0409	-0.1498	0.0449	-0.0026	-1	0.1981	0.0653	-0.1548	0.0254	-0.0065
0	0.4468	0.0477	-0.1477	0.0544	-0.0035	0	0.3036	0.0711	-0.1778	0.0377	-0.0071
1	0.5421	0.0577	-0.1477	0.0476	-0.0045	1	0.4236	0.0818	-0.1989	0.0514	-0.0088
2	0.6364	0.0668	-0.1436	0.0761	-0.0058	2	0.5104	0.0935	-0.2104	0.0619	-0.0103
3	0.7160	0.0786	-0.1346	0.0853	-0.0072	3	0.6130	0.1099	-0.2264	0.0746	-0.0120
5	0.8529	0.1123	-0.1334	0.1033	-0.0115	5	0.7634	0.1450	-0.2424	0.0912	-0.0166
7	0.9805	0.1560	-0.1407	0.1203	-0.0168	7	0.9166	0.1976	-0.2584	0.1095	-0.0227
10	1.0915	0.2298	-0.1518	0.1370	-0.0267	10	1.0843	0.2787	-0.2820	0.1291	-0.0340
15	0.9324	0.3230	-0.1981	0.1167	-0.0387	15	1.3012	0.4508	-0.3288	0.1579	-0.0279
20	0.9620	0.4285	-0.2177	0.1172	-0.0414						
25	0.9435	0.5332	-0.2251	0.1150	-0.0324						
$M = 0.90$											
$M = 1.10$											
-10	-0.4884	0.1172	-0.0706	-0.0599	-0.0157	-10	-0.5055	0.1294	-0.0248	-0.0585	-0.0175
-7	-0.3206	0.0715	-0.0979	-0.0359	-0.0087	-7	-0.3158	0.1032	-0.0640	-0.0348	-0.0114
-5	-0.1233	0.0477	-0.1306	-0.0120	-0.0045	-5	-0.1731	0.0698	-0.0901	-0.0177	-0.0083
-3	0.1019	0.0416	-0.1626	0.0140	-0.0029	-3	0.0028	0.0607	-0.1207	0.0029	-0.0066
-2	0.2138	0.0424	-0.1688	0.0269	-0.0032	-2	0.0900	0.0596	-0.1330	0.0135	-0.0064
-1	0.3174	0.0485	-0.1688	0.0385	-0.0043	-1	0.1842	0.0633	-0.1474	0.0237	-0.0067
0	0.4061	0.0554	-0.1706	0.0493	-0.0050	0	0.2839	0.0698	-0.1679	0.0361	-0.0073
1	0.4982	0.0668	-0.1713	0.0603	-0.0062	1	0.3809	0.0789	-0.1832	0.0471	-0.0085
2	0.6002	0.0809	-0.1778	0.0716	-0.0080	2	0.4778	0.0913	-0.1992	0.0583	-0.0097
3	0.6824	0.0950	-0.1876	0.0828	-0.0102	3	0.5678	0.1046	-0.2126	0.0689	-0.0115
5	0.8386	0.1314	-0.1971	0.1014	-0.0151	5	0.7257	0.1403	-0.2322	0.0874	-0.0159
7	0.9669	0.1835	-0.2139	0.1185	-0.0226	7	0.8670	0.1907	-0.2487	0.1042	-0.0218
10	1.0107	0.2619	-0.2248	0.1377	-0.0325	10	1.0304	0.2670	-0.2708	0.1230	-0.0325
15	1.0162	0.3453	-0.2284	0.1253	-0.0435	15	1.2548	0.4386	-0.3223	0.1513	-0.0541
20	1.0491	0.4690	-0.2473	0.1277	-0.0587						
25	1.0326	0.5726	-0.2539	0.1253	-0.0723						

TABLE 4.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 4 MODEL - Concluded

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-1719					-10	-4769	.1280	-0.0602	-0.0543	-0.0180
-7	.0081					-7	-2533	.0857	-0.1121	-0.0254	-0.0117
-5	.1478					-5	-0547	.0696	-0.1557	-0.0025	-0.0080
-3	.2767					-3	.1845	.0685	-0.1971	.0245	-0.0070
-2	.3439					-2	.3064	.0727	-0.2109	.0381	-0.0073
-1	.4057					-1	.4096	.0804	-0.2213	.0495	-0.0083
0	.4513					0	.5253	.0915	-0.2241	.0607	-0.0095
1	.5131					1	.5957	.1049	-0.2331	.0713	-0.0109
2	.5668					2	.6895	.1199	-0.2331	.0818	-0.0126
3	.6260					3	.7755	.1391	-0.2414	.0922	-0.0148
5	.7415					5	.9256	.1810	-0.2608	.1097	-0.0199
7	.8462					7	1.0725	.2384	-0.2712	.1249	-0.0265
10	.9053					10	1.1882	.3175	-0.2850	.1408	-0.0364
15	.8624					15	1.3665	.4783	-0.2988	.1643	-0.0573
20	.8465					20	1.1976	.5505	-0.2884	.1408	-0.0658
25	.8113					25	1.1194	.6351	-0.2760	.1302	-0.0767
$M = 0.60$											
$M = 1.00$											
-10	-2045	.0513	-0.0861	-0.0248	-0.0054	-10	-4834	.1237	-0.0525	-0.0521	-0.0172
-7	.0206	.0189	-0.1182	.0035	-0.0018	-7	-2417	.0868	-0.1036	-0.0253	-0.0112
-5	.1659	.0189	-0.1206	.0203	-0.0010	-5	-0826	.0739	-0.1388	-0.0062	-0.0087
-3	.3166	.0255	-0.1225	.0376	-0.0008	-3	.1066	.0706	-0.1707	.0153	-0.0076
-2	.3989	.0291	-0.1213	.0459	-0.0010	-2	.2417	.0739	-0.1940	.0299	-0.0076
-1	.4510	.0365	-0.1201	.0539	-0.0016	-1	.3678	.0805	-0.2125	.0445	-0.0081
0	.5085	.0430	-0.1177	.0607	-0.0019	0	.4624	.0898	-0.2205	.0554	-0.0093
1	.5730	.0526	-0.1170	.0689	-0.0022	1	.5600	.1027	-0.2251	.0663	-0.0107
2	.6456	.0628	-0.1140	.0769	-0.0028	2	.6425	.1174	-0.2305	.0763	-0.0123
3	.7210	.0762	-0.1110	.0852	-0.0042	3	.7296	.1351	-0.2371	.0864	-0.0143
5	.8307	.1031	-0.1074	.0985	-0.0073	5	.8797	.1776	-0.2544	.1035	-0.0196
7	.9294	.1516	-0.1153	.1118	-0.0128	7	1.0058	.2348	-0.2756	.1188	-0.0259
10	.9540	.2245	-0.1529	.1184	-0.0223	10	1.1650	.3226	-0.3002	.1381	-0.0365
15	.9074	.3208	-0.1826	.1138	-0.0362	15	1.3601	.4917	-0.3175	.1625	-0.0581
20	.8937	.4153	-0.1886	.1095	-0.0478	20	1.4202	.6541	-0.3281	.1680	-0.0777
25	.8855	.5151	-0.2001	.1075	-0.0598	25	1.3151	.7707	-0.3387	.1552	-0.0943
$M = 0.80$											
$M = 1.05$											
-10	-2529	.0697	-0.1176	-0.0283	-0.0084	-10	-4456	.1199	-0.0480	-0.0507	-0.0175
-7	-0.0491	.0374	-0.1505	.0030	-0.0039	-7	-2488	.0839	-0.0954	-0.0507	-0.0113
-5	.1204	.0324	-0.1537	.0159	-0.0024	-5	-0926	.0722	-0.1293	-0.0077	-0.0087
-3	.2761	.0400	-0.1525	.0341	-0.0024	-3	.0810	.0680	-0.1581	.0121	-0.0080
-2	.3484	.0456	-0.1525	.0430	-0.0032	-2	.1924	.0705	-0.1773	.0248	-0.0081
-1	.4290	.0524	-0.1508	.0524	-0.0040	-1	.3110	.0765	-0.1965	.0379	-0.0086
0	.5152	.0619	-0.1517	.0616	-0.0051	0	.4297	.0858	-0.2125	.0520	-0.0097
1	.5874	.0702	-0.1463	.0699	-0.0058	1	.5281	.0982	-0.2208	.0630	-0.0109
2	.6504	.0775	-0.1345	.0778	-0.0066	2	.6149	.1128	-0.2278	.0727	-0.0125
3	.7468	.0930	-0.1340	.0891	-0.0082	3	.7017	.1309	-0.2349	.0827	-0.0146
5	.8858	.1268	-0.1369	.1077	-0.0115	5	.8478	.1707	-0.2528	.0997	-0.0194
7	1.0118	.1768	-0.1463	.1241	-0.0174	7	.9722	.2248	-0.2707	.1148	-0.0130
10	1.1007	.2533	-0.1565	.1376	-0.0269	10	1.1313	.3138	-0.2957	.1338	-0.0184
15	.9599	.3417	-0.1975	.1174	-0.0393	15	1.3541	.4923	-0.3380	.1608	-0.0589
20	.9673	.4466	-0.2107	.1196	-0.1039	20	1.4988	.6759	-0.3546	.1784	-0.0824
25	.9488	.5504	-0.2205	.1138	-0.0651	25	1.3946	.7912	-0.3584	.1643	-0.0993
$M = 0.90$											
$M = 1.10$											
-10	-3905	.1065	-0.0991	-0.0454	-0.0151	-10	-4324	.1193	-0.0460	-0.0488	-0.0168
-7	-2.009	.0713	-0.1319	-0.0194	-0.0078	-7	-2453	.0818	-0.0901	-0.0259	-0.0115
-5	.0165	.0547	-0.1683	.0054	-0.0053	-5	-0970	.0703	-0.1202	-0.0086	-0.0089
-3	.2371	.0583	-0.1909	.0305	-0.0055	-3	.0804	.0668	-0.1533	.0113	-0.0081
-2	.3342	.0644	-0.1930	.0416	-0.0060	-2	.1885	.0771	-0.1711	.0230	-0.0084
-1	.4083	.0685	-0.1872	.0502	-0.0071	-1	.2827	.0750	-0.1870	.0346	-0.0090
0	.5071	.0822	-0.1945	.0611	-0.0082	0	.3950	.0825	-0.2023	.0476	-0.0101
1	.5894	.0924	-0.1909	.0711	-0.0095	1	.4989	.0941	-0.2128	.0585	-0.0114
2	.6767	.1044	-0.1887	.0811	-0.0109	2	.5834	.1091	-0.2201	.0691	-0.0130
3	.7525	.1207	-0.1981	.0973	-0.0127	3	.6652	.1247	-0.2262	.0784	-0.0151
5	.8759	.1551	-0.1981	.1063	-0.0168	5	.8038	.1621	-0.2404	.0949	-0.0201
7	1.0044	.2114	-0.2164	.1223	-0.0232	7	.9313	.2133	-0.2606	.1097	-0.0269
10	1.1394	.2939	-0.2382	.1403	-0.0337	10	1.0810	.2985	-0.2881	.1278	-0.0367
15	1.0439	.2923	-0.2309	.1251	-0.0440	15	1.2916	.4676	-0.3274	.1544	-0.0588
20	1.0636	.4987	-0.2491	.1275	-0.0594	20	1.4579	.6652	-0.3630	.1742	-0.0864
25	1.0241	.5992	-0.2513	.1227	-0.0722	25	1.4025	.6925	-0.4745	.1722	-0.1054

TABLE 5.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 5 MODEL

 $\frac{t}{c} = 0.06$        $\frac{c_f}{c} = \text{NONE}$ 

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.50$											
-10	-0.5664					-10	-0.9107	0.1824	0.1735	-0.1116	-0.0280
-7	-0.4377					-7	-0.6750	0.1086	0.1092	-0.0824	-0.0169
-5	-0.3175					-5	-0.5062	0.0690	0.0746	-0.0619	-0.0115
-3	-0.2060					-3	-0.3251	0.0458	0.0472	-0.0405	-0.0069
-2	-0.1459					-2	-0.2333	0.0378	0.0335	-0.0294	-0.0054
-1	-0.0858					-1	-0.1290	0.0299	0.0159	-0.0161	-0.0039
0	-0.0172					0	-0.0223	0.0354	0.0088	-0.0043	-0.0027
1	0.0472					1	0.0819	0.0335	0.0005	0.0080	-0.0024
2	0.1159					2	0.1762	0.0385	-0.0082	0.0186	-0.0025
3	0.1802					3	0.2730	0.0458	-0.0214	0.0304	-0.0032
5	0.3004					5	0.4467	0.0720	-0.0488	0.0508	-0.0057
7	0.4334					7	0.6328	0.1117	-0.0884	0.0730	-0.0098
10	0.6179					10	0.8958	0.1941	-0.1641	0.1031	-0.0192
15	0.6995					15	1.0812	0.3576	-0.2135	0.1407	-0.0386
20	0.7381										
25	0.6995										
$M = 0.60$											
$M = 0.70$											
-10	-0.6817	0.1286	0.0574	-0.0860	-0.0199	-10	-0.8492	0.1732	0.1605	-0.1044	-0.0261
-7	-0.5629	0.0649	0.0019	-0.0673	-0.0112	-7	-0.6351	0.1047	0.1016	-0.0776	-0.0161
-5	-0.4200	0.0334	-0.0010	-0.0493	-0.0066	-5	-0.4805	0.0708	0.0742	-0.0587	-0.0116
-3	-0.2617	0.0163	0.0005	-0.0303	-0.0034	-3	-0.3069	0.0485	0.0505	-0.0386	-0.0074
-2	-0.1781	0.0152	0.0005	-0.0207	-0.0026	-2	-0.2141	0.0403	0.0358	-0.0270	-0.0058
-1	-0.1012	0.0141	-0.0015	-0.0113	-0.0016	-1	-0.1189	0.0381	0.0216	-0.0155	-0.0047
0	-0.0220	0.0141	0.0005	-0.0024	-0.0008	0	-0.0071	0.0387	0.0084	-0.0035	-0.0034
1	0.0638	0.0141	0.0010	-0.0073	-0.0005	1	0.0880	0.0409	-0.0058	0.0079	-0.0034
2	0.1451	0.0163	0.0025	0.0164	-0.0006	2	0.1879	0.0467	-0.0195	0.0187	-0.0038
3	0.2221	0.0205	0.0019	0.0254	-0.0010	3	0.2759	0.0544	-0.0321	0.0298	-0.0043
5	0.3760	0.0334	0.0039	0.0438	-0.0019	5	0.4377	0.0778	-0.0600	0.0494	-0.0066
7	0.5431	0.0638	0.0054	0.0626	-0.0031	7	0.6137	0.1158	-0.0931	0.0702	-0.0107
10	0.7147	0.1308	-0.0326	0.0850	-0.0105	10	0.8302	0.1842	-0.1426	0.0958	-0.0193
15	0.7806	0.2335	-0.0988	0.0997	-0.0248	15	1.1798	0.3544	-0.2205	0.1372	-0.0394
20	0.7564	0.3094	-0.1221	0.0965	-0.0359						
25	0.7655	0.4013	-0.1392	0.0965	-0.0466						
$M = 0.80$											
$M = 0.90$											
-10	-0.7929	0.1341	0.0427	-0.0999	-0.0213	-10	-0.8231	0.1687	0.1483	-0.0992	-0.0251
-7	-0.6514	0.0718	0.0134	-0.0804	-0.0123	-7	-0.6019	0.1015	0.0974	-0.0737	-0.0158
-5	-0.5129	0.0385	-0.0019	-0.0607	-0.0074	-5	-0.4537	0.0678	0.0721	-0.0558	-0.0114
-3	-0.3066	0.0167	0.0085	-0.0358	-0.0037	-3	-0.2896	0.0465	0.0469	-0.0361	-0.0073
-2	-0.2181	0.0116	-0.0045	-0.0249	-0.0025	-2	-0.1984	0.0398	0.0333	-0.0255	-0.0060
-1	-0.1267	0.0116	-0.0017	-0.0146	-0.0016	-1	-0.1094	0.0381	0.0217	-0.0151	-0.0047
0	-0.0354	0.0116	0.0013	-0.0034	-0.0009	0	-0.0091	0.0381	0.0081	-0.0038	-0.0036
1	0.0796	0.0131	0.0006	0.0092	-0.0006	1	0.0889	0.0409	-0.0071	0.0080	-0.0036
2	0.1680	0.0167	0.0032	0.0195	-0.0006	2	0.1756	0.0454	-0.0187	0.0179	-0.0037
3	0.2594	0.0218	0.0085	0.0295	-0.0010	3	0.2645	0.0532	-0.0313	0.0283	-0.0044
5	0.4480	0.0427	0.0098	0.0521	-0.0020	5	0.4195	0.0746	-0.0570	0.0467	-0.0065
7	0.6219	0.0797	-0.0045	0.0736	-0.0049	7	0.5837	0.1110	-0.0867	0.0666	-0.0106
10	0.7841	0.1435	-0.0313	0.0953	-0.0119	10	0.7912	0.1783	-0.1367	0.0910	-0.0179
15	0.8135	0.2464	-0.1056	0.1022	-0.0257	15	1.1218	0.3397	-0.2224	0.1306	-0.0370
20	0.8047	0.3319	-0.1392	0.1002	-0.0370						
25	0.8430	0.4364	-0.1591	0.1033	-0.0499						
$M = 0.90$											
$M = 1.00$											
-10	-0.8492	0.1732	0.1605	-0.1044	-0.0261	-10	-0.8231	0.1687	0.1483	-0.0992	-0.0251
-7	-0.6351	0.1047	0.1016	-0.0776	-0.0161	-7	-0.6019	0.1015	0.0974	-0.0737	-0.0158
-5	-0.4805	0.0708	0.0721	-0.0558	-0.0114	-5	-0.4537	0.0678	0.0721	-0.0536	-0.0114
-3	-0.3069	0.0485	0.0505	-0.0386	-0.0074	-3	-0.2896	0.0465	0.0469	-0.0361	-0.0073
-2	-0.2141	0.0403	0.0358	-0.0270	-0.0058	-2	-0.1984	0.0426	0.0451	-0.0347	-0.0074
-1	-0.1189	0.0381	0.0216	-0.0155	-0.0047	-1	-0.1094	0.0381	0.0217	-0.0151	-0.0047
0	-0.0071	0.0387	0.0084	-0.0035	-0.0034	0	-0.0091	0.0381	0.0081	-0.0038	-0.0036
1	0.0880	0.0409	-0.0071	0.0080	-0.0036	1	0.0889	0.0409	-0.0071	0.0080	-0.0036
2	0.1879	0.0467	-0.0195	0.0187	-0.0037	2	0.1756	0.0454	-0.0187	0.0179	-0.0037
3	0.2645	0.0532	-0.0313	0.0283	-0.0044	3	0.2645	0.0532	-0.0313	0.0283	-0.0044
5	0.4195	0.0746	-0.0570	0.0467	-0.0065	5	0.4195	0.0746	-0.0570	0.0467	-0.0065
7	0.5837	0.1110	-0.0867	0.0666	-0.0106	7	0.5837	0.1110	-0.0867	0.0666	-0.0106
10	0.7912	0.1783	-0.1367	0.0910	-0.0179	10	0.7912	0.1783	-0.1367	0.0910	-0.0179
15	1.1218	0.3397	-0.2224	0.1306	-0.0370	15	1.1218	0.3397	-0.2224	0.1306	-0.0370
$M = 1.10$											
-10	-0.7783	0.1622	0.1436	-0.0956	-0.0247	-10	-0.7584	0.0959	0.0975	-0.0717	-0.0158
-7	-0.6584	0.0959	0.0975	-0.0717	-0.0158	-7	-0.4407	0.0636	0.0708	-0.0543	-0.0117
-5	-0.4407	0.0636	0.0721	-0.0558	-0.0114	-5	-0.2784	0.0426	0.0451	-0.0347	-0.0074
-3	-0.2784	0.0426	0.0451	-0.0347	-0.0074	-3	-0.1885	0.0366	0.0320	-0.0245	-0.0059
-2	-0.1885	0.0366	0.0320	-0.0245	-0.0059	-2	-0.1099	0.0356	0.0209	-0.0138	-0.0048
-1	-0.1099	0.0356	0.0209	-0.0138	-0.0048	-1	-0.0088	0.0362	0.0078	-0.0040	-0.0038
0	-0.0088	0.0362	0.0078	-0.0040	-0.0038	0	-0.0088	0.0366	-0.0053	0.0070	-0.0040
1	0.0833	0.0366	-0.0053	0.0070	-0.0040	1	0.0833	0.0366	-0.0053	0.0070	-0.0040
2	0.1732	0.0410	-0.0179	0.0172	-0.0044	2	0.1732	0.0410	-0.0179	0.0172	-0.0044
3	0.2543	0.0486	-0.0296	0.0273	-0.0050	3	0.2543	0.0486	-0.0296	0.0273	-0.0050
5	0.4056	0.0700	-0.0543	0.0451	-0.0071	5	0.4056	0.0700	-0.0543	0.0451	-0.0071
7	0.5656	0.1046	-0.0854	0.0643	-0.0112	7	0.5656	0.1046	-0.0854	0.0643	-0.0112
10	0.7673	0.1714	-0.1329	0.0881	-0.0183	10	0.7673	0.1714	-0.1329	0.0881	-0.0183
15	1.0655	0.3246	-0.2071	0.1256	-0.0366	15	1.0655	0.3246	-0.2071	0.1256	-0.0366

TABLE 5.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 5 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{C_f}{c} = 0.20$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.50$											
-10	-0.3101					-10	-0.7009	0.1559	0.0342	-0.0825	-0.0225
-7	-0.1240					-7	-0.4904	0.0962	-0.0115	-0.0558	-0.0134
-5	0.0043					-5	-0.3220	0.0718	-0.0430	-0.0361	-0.0096
-3	0.1433					-3	-0.1684	0.0476	-0.0567	-0.0168	-0.0052
-2	0.2181					-2	-0.0545	0.0445	-0.0813	-0.0031	-0.0040
-1	0.2823					-1	0.0842	0.0445	-0.1169	0.0127	-0.0040
0	0.3425					0	0.2452	0.0530	-0.1572	0.0313	-0.0047
1	0.4045					1	0.3888	0.0670	-0.1874	0.0474	-0.0058
2	0.4534					2	0.5077	0.0780	-0.2038	0.0618	-0.0071
3	0.5132					3	0.6241	0.0962	-0.2202	0.0757	-0.0090
5	0.6351					5	0.7851	0.1327	-0.2416	0.0945	-0.0132
7	0.7635					7	0.9461	0.1851	-0.2662	0.1130	-0.0190
10	0.8447					10	1.1318	0.2759	-0.2969	0.1371	-0.0295
15	0.8640										
20	0.8383										
25	0.7934										
$M = 0.60$											
$M = 1.00$											
-10	-0.3585	0.0798	-0.0665	-0.0431	-0.0104	-10	-0.6785	0.1476	0.0362	-0.0790	-0.0215
-7	-0.1535	0.0248	-0.1271	-0.0160	-0.0028	-7	-0.4694	0.0951	-0.0129	-0.0528	-0.0138
-5	0.0022	0.0125	-0.1285	0.0024	-0.0014	-5	-0.3060	0.0718	-0.0451	-0.0334	-0.0098
-3	0.1513	0.0145	-0.1271	0.0195	-0.0011	-3	-0.1423	0.0590	-0.0714	-0.0138	-0.0067
-2	0.2214	0.0162	-0.1223	0.0271	-0.0008	-2	-0.0427	0.0554	-0.0890	-0.0016	-0.0059
-1	0.2850	0.0204	-0.1173	0.0341	-0.0010	-1	0.0641	0.0250	-0.1081	0.0104	-0.0052
0	0.3639	0.0259	-0.1183	0.0433	-0.0012	0	0.1756	0.0590	-0.1270	0.0230	-0.0052
1	0.4758	0.0340	-0.1271	0.0577	-0.0013	1	0.3013	0.0648	-0.1516	0.0362	-0.0057
2	0.5393	0.0421	-0.1173	0.0654	-0.0015	2	0.4318	0.0764	-0.1821	0.0537	-0.0070
3	0.6227	0.0528	-0.1150	0.0741	-0.0023	3	0.5623	0.0922	-0.2083	0.0684	-0.0088
5	0.7586	0.0787	-0.1081	0.0900	-0.0048	5	0.7354	0.1278	-0.2330	0.0887	-0.0129
7	0.8814	0.1201	-0.1077	0.1054	-0.0101	7	0.8944	0.1797	-0.2550	0.1071	-0.0187
10	0.9365	0.1903	-0.1562	0.1171	-0.0191	10	1.0771	0.2655	-0.2855	0.1313	-0.0285
15	0.9230	0.2944	-0.1867	0.1213	-0.0338						
20	0.8901	0.3806	-0.1886	0.1141	-0.0454						
$M = 0.80$											
$M = 1.05$											
-10	-0.4898	0.0944	-0.0738	-0.0593	-0.0137	-10	-0.6440	0.1427	0.0335	-0.0751	-0.0212
-7	-0.3383	0.0459	-0.1080	-0.0386	-0.0066	-7	-0.4460	0.0912	-0.0106	-0.0508	-0.0138
-5	-0.1515	0.0238	-0.1253	-0.0164	-0.0026	-5	-0.2981	0.0700	-0.0400	-0.0331	-0.0097
-3	0.0677	0.0181	-0.1314	0.0090	-0.0011	-3	-0.1320	0.0577	-0.0684	-0.0133	-0.0069
-2	0.1721	0.0210	-0.1340	0.0207	-0.0010	-2	-0.0410	0.0537	-0.0848	-0.0022	-0.0059
-1	0.2751	0.0257	-0.1340	0.0329	-0.0012	-1	0.0614	0.0520	-0.1012	0.0095	-0.0058
0	0.3765	0.0310	-0.1350	0.0447	-0.0019	0	0.1661	0.0560	-0.1206	0.0219	-0.0058
1	0.4810	0.0402	-0.1422	0.0571	-0.0025	1	0.2731	0.0633	-0.1404	0.0345	-0.0065
2	0.6089	0.0499	-0.1454	0.0731	-0.0038	2	0.3800	0.0717	-0.1621	0.0471	-0.0075
3	0.7060	0.0632	-0.1416	0.0848	-0.0048	3	0.5097	0.1432	-0.1888	0.0619	-0.0091
5	0.8649	0.0980	-0.1497	0.1057	-0.0085	5	0.6895	0.1220	-0.2235	0.0833	-0.0130
7	0.9737	0.1440	-0.1568	0.1214	-0.0142	7	0.8510	0.1707	-0.2436	0.1016	-0.0185
10	1.0737	0.2177	-0.1584	0.1354	-0.0236	10	1.0285	0.2535	-0.2748	0.1250	-0.0281
15	0.9237	0.3009	-0.2017	0.1171	-0.0356						
$M = 0.90$											
$M = 1.10$											
-10	-0.6772	0.1397	-0.0051	-0.0835	-0.0206	-10	-0.6194	0.1389	0.0358	-0.0727	-0.0213
-7	-0.5051	0.0812	-0.0233	-0.0596	-0.0119	-7	-0.4334	0.0899	-0.0102	-0.0510	-0.0135
-5	-0.3495	0.0524	-0.0519	-0.0384	-0.0075	-5	-0.2911	0.0673	-0.0361	-0.0321	-0.0100
-3	-0.0983	0.0343	-0.1093	-0.0072	-0.0032	-3	-0.1226	0.0555	-0.0658	-0.0121	-0.0071
-2	0.0464	0.0336	-0.1335	0.0353	-0.0025	-2	-0.0350	0.0517	-0.0811	-0.0021	-0.0063
-1	0.1966	0.0375	-0.1613	0.0260	-0.0025	-1	0.0613	0.0511	-0.0981	0.0091	-0.0059
0	0.3195	0.0456	-0.1673	0.0398	-0.0029	0	0.1576	0.0544	-0.1155	0.0202	-0.0060
1	0.4369	0.0557	-0.1764	0.0533	-0.0041	1	0.2670	0.0608	-0.1365	0.0329	-0.0065
2	0.5516	0.0692	-0.1824	0.0668	-0.0057	2	0.3655	0.0705	-0.1544	0.0446	-0.0074
3	0.6608	0.0860	-0.1915	0.0795	-0.0074	3	0.4772	0.0850	-0.1767	0.0574	-0.0088
5	0.8328	0.1256	-0.2096	0.1007	-0.0121	5	0.6610	0.1173	-0.2140	0.0786	-0.0127
7	0.9966	0.1834	-0.2398	0.1206	-0.0187	7	0.8164	0.1642	-0.2363	0.0971	-0.0178
10	1.1796	0.2759	-0.2670	0.1447	-0.0295	10	0.9937	0.2449	-0.2634	0.1199	-0.0272
15	1.3762	0.2135	-0.2730	0.1670	-0.0493						
20	1.1195	0.2384	-0.2609	0.1352	-0.0554						

TABLE 5.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 5 MODEL - Continued

$$\frac{c_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-.2299					-10	-.6020	.1420	-.0151	-.0722	-.0206
-7	-.0434					-7	-.3815	.0927	-.0636	-.0440	-.0128
-5	.0954					-5	-.2180	.0701	-.0951	-.0233	-.0087
-3	.2342					-3	-.0198	.0555	-.1293	-.0000	-.0061
-2	.3036					-2	.1338	.0578	-.1655	.0176	-.0062
-1	.3774					-1	.2899	.0639	-.1929	.0349	-.0062
0	.4294					0	.4385	.0725	-.2093	.0493	-.0074
1	.4858					1	.5351	.0853	-.2176	.0637	-.0088
2	.5422					2	.6243	.1005	-.2247	.0741	-.0103
3	.6029					3	.7259	.1207	-.2340	.0854	-.0125
5	.7243					5	.8721	.1614	-.2493	.1027	-.0168
7	.8458					7	1.0133	.2163	-.2669	.1188	-.0232
10	.9152					10	1.1817	.3065	-.2871	.1419	-.0338
$M = 0.60$											
-10	-.2850	.0702	-.0776	-.0349	-.0085	-10	-.5723	.1373	-.0123	-.0692	-.0198
-7	-.0614	.0259	-.1256	-.0060	-.0014	-7	-.3633	.0893	-.0583	-.0420	-.0128
-5	.1030	.0162	-.1256	.0134	-.0010	-5	-.2019	.0706	-.0903	-.0224	-.0089
-3	.2784	.0195	-.1280	.0332	-.0010	-3	-.0309	.0625	-.1195	-.0012	-.0071
-2	.3596	.0237	-.1295	.0426	-.0012	-2	.0760	.0613	-.1376	.0106	-.0066
-1	.4429	.0303	-.1295	.0515	-.0017	-1	.1995	.0642	-.1586	.0242	-.0065
0	.5152	.0344	-.1271	.0607	-.0025	0	.3420	.0718	-.1875	.0415	-.0072
1	.5635	.0410	-.1183	.0668	-.0029	1	.4916	.0835	-.2085	.0572	-.0087
2	.6405	.0506	-.1150	.0754	-.0038	2	.5747	.0963	-.2190	.0685	-.0102
3	.7104	.0625	-.1101	.0834	-.0049	3	.6744	.1168	-.2269	.0795	-.0123
5	.8551	.0895	-.1028	.1007	-.0082	5	.8240	.1547	-.2400	.0968	-.0165
7	.9406	.1348	-.1077	.1145	-.0136	7	.9546	.2073	-.2553	.1130	-.0224
10	.9757	.2081	-.1450	.1222	-.0234	10	1.1280	.2960	-.2826	.1360	-.0328
$M = 0.80$											
-10	-.3824	.0833	-.1103	-.0446	-.0116	-10	-.5465	.1343	-.0126	-.0652	-.0192
-7	-.1956	.0427	-.1406	-.0210	-.0056	-7	-.3529	.0879	-.0539	-.0409	-.0126
-5	-.0059	.0275	-.1506	.0099	-.0025	-5	-.2004	.0689	-.0816	-.0223	-.0095
-3	.1839	.0290	-.1461	.0223	-.0023	-3	-.0296	.0616	-.1118	-.0022	-.0071
-2	.2780	.0318	-.1451	.0331	-.0025	-2	.0706	.0587	-.1254	.0097	-.0070
-1	.3663	.0377	-.1419	.0436	-.0032	-1	.1708	.0633	-.1435	.0210	-.0067
0	.4619	.0449	-.1412	.0546	-.0040	0	.2892	.0689	-.1647	.0347	-.0073
1	.5545	.0543	-.1431	.0657	-.0049	1	.4190	.0789	-.1899	.0497	-.0086
2	.6725	.0672	-.1454	.0805	-.0062	2	.5283	.0923	-.2050	.0630	-.0098
3	.7707	.0796	-.1357	.0923	-.0075	3	.6239	.1092	-.2176	.0741	-.0115
5	.9149	.1172	-.1448	.1105	-.0112	5	.7788	.1462	-.2317	.0913	-.0155
7	1.0179	.1664	-.1503	.1254	-.0169	7	.9154	.1959	-.2463	.1072	-.0213
10	1.1179	.2481	-.1568	.1385	-.0265	10	1.0770	.2822	-.2730	.1296	-.0311
$M = 0.90$											
-10	-.5639	.1330	-.0595	-.0703	-.0189	-10	-.5279	.1281	-.0106	-.0629	-.0185
-7	-.4345	.0826	-.0640	-.0504	-.0117	-7	-.3417	.0845	-.0494	-.0396	-.0124
-5	-.2225	.0537	-.1048	-.0239	-.0069	-5	-.2037	.0657	-.0760	-.0223	-.0093
-3	.0642	.0437	-.1613	.0093	-.0041	-3	-.0350	.0592	-.1051	-.0021	-.0074
-2	.2075	.0470	-.1782	.0260	-.0043	-2	.0591	.0582	-.1202	.0085	-.0068
-1	.3372	.0571	-.1945	.0403	-.0053	-1	.1533	.0609	-.1366	.0202	-.0067
0	.4533	.0605	-.1885	.0517	-.0057	0	.2628	.0663	-.1526	.0315	-.0076
1	.5488	.0759	-.2005	.0650	-.0072	1	.3811	.1034	-.1754	.0453	-.0084
2	.6444	.0886	-.1975	.0769	-.0088	2	.4928	.0888	-.1948	.0585	-.0098
3	.7485	.1095	-.2066	.0901	-.0116	3	.5957	.1050	-.2093	.0708	-.0119
5	.9205	.1859	-.2277	.1100	-.0170	5	.7425	.1416	-.2234	.0872	-.0153
7	1.0567	.2082	-.2398	.1267	-.0232	7	.8805	.1901	-.2384	.1029	-.0207
10	1.2260	.2989	-.2609	.1511	-.0170	10	1.0338	.2730	-.2636	.1238	-.0300
$M = 1.00$											
-10	-.5723	.1373	-.0123	-.0692	-.0198	-10	-.5465	.1343	-.0126	-.0652	-.0192
-7	-.3633	.0893	-.0583	-.0420	-.0128	-7	-.3529	.0879	-.0539	-.0409	-.0126
-5	-.2019	.0706	-.0903	-.0224	-.0089	-5	-.2004	.0689	-.0816	-.0223	-.0095
-3	-.0309	.0625	-.1195	-.0012	-.0071	-3	-.0296	.0616	-.1118	-.0022	-.0071
-2	.0760	.0613	-.1376	.0106	-.0066	-2	.0706	.0587	-.1254	.0097	-.0070
-1	.1995	.0642	-.1586	.0242	-.0065	-1	.1708	.0633	-.1435	.0210	-.0067
0	.3420	.0718	-.1875	.0415	-.0072	0	.2892	.0689	-.1647	.0347	-.0073
1	.4916	.0835	-.2085	.0572	-.0087	1	.4190	.0789	-.1899	.0497	-.0086
2	.5747	.0963	-.2190	.0685	-.0102	2	.5283	.0923	-.2050	.0630	-.0098
3	.6744	.1168	-.2269	.0795	-.0123	3	.6239	.1092	-.2176	.0741	-.0115
5	.8240	.1547	-.2400	.0968	-.0165	5	.7788	.1462	-.2317	.0913	-.0155
7	.9546	.2073	-.2553	.1130	-.0224	7	.9154	.1959	-.2463	.1072	-.0213
10	1.1280	.2960	-.2826	.1360	-.0328	10	1.0770	.2822	-.2730	.1296	-.0311
$M = 1.05$											
-10	-.5465	.1343	-.0126	-.0652	-.0192	-10	-.5279	.1281	-.0106	-.0629	-.0185
-7	-.3529	.0879	-.0539	-.0409	-.0126	-7	-.3417	.0845	-.0494	-.0396	-.0124
-5	-.2004	.0689	-.0816	-.0223	-.0095	-5	-.2037	.0657	-.0760	-.0223	-.0093
-3	-.0296	.0616	-.1118	-.0022	-.0071	-3	-.0350	.0592	-.1051	-.0021	-.0074
-2	.0706	.0587	-.1254	.0097	-.0066	-2	.0591	.0582	-.1202	.0085	-.0068
-1	.1708	.0633	-.1435	.0210	-.0067	-1	.1533	.0609	-.1366	.0202	-.0067
0	.2892	.0689	-.1647	.0347	-.0073	0	.2628	.0663	-.1526	.0315	-.0076
1	.4190	.0789	-.1899	.0497	-.0086	1	.3811	.1034	-.1754	.0453	-.0084
2	.5283	.0923	-.2050	.0630	-.0098	2	.4928	.0888	-.1948	.0585	-.0098
3	.6239	.1092	-.2176	.0741	-.0115	3	.5957	.1050	-.2093	.0708	-.0119
5	.7788	.1462	-.2317	.0913	-.0155	5	.7425	.1416	-.2234	.0872	-.0153
7	.9154	.1959	-.2463	.1072	-.0213	7	.8805	.1901	-.2384	.1029	-.0207
10	1.0770	.2822	-.2730	.1296	-.0311	10	1.0338	.2730	-.2636	.1238	-.0300
$M = 1.10$											
-10	-.5279	.1281	-.0106	-.0629	-.0185	-10	-.5129	.1251	-.0106	-.0629	-.0185
-7	-.3417	.0845	-.0494	-.0396	-.0124	-7	-.3317	.0815	-.0494	-.0396	-.0124
-5	-.2037	.0657	-.0760	-.0223	-.0093	-5	-.2137	.0687	-.0760	-.0223	-.0093
-3	-.0350	.0592	-.1051	-.0021	-.0074	-3	-.0450	.0592	-.1051	-.0021	-.0074
-2	.0591	.0582	-.1202	.0085	-.0068	-2	.0513	.0582	-.1202	.0085	-.0068
-1	.1533	.0609	-.1366	.0202	-.0067	-1	.1411	.0609	-.1366	.0202	-.0067
0	.2628	.0663	-.1526	.0315	-.0076	0	.2448	.0663	-.1526	.0315	-.0076
1	.3811	.1034	-.1754	.0453	-.0084	1	.3561	.1034	-.1754	.0453	-.0084
2	.4928	.0888	-.1948	.0585	-.0098	2	.4688	.0888	-.1948	.0585	-.0098
3	.5957	.1050	-.2093	.0708	-.0119	3	.5688	.1050	-.2093	.0708	-.0119
5	.7425	.1416	-.2234	.0872	-.0153	5	.7145	.1416	-.2234	.0872	-.0153
7	.8805	.1901	-.2384	.1029	-.0207	7	.8522	.1901	-.2384	.1029	-.0207
10	1.0338	.2730	-.2636	.1238	-.0300	10	1.0055	.2730	-.2636	.1238	-.0300

TABLE 5.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 5 MODEL - Concluded

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.95$											
-10	-0.2035					-10	-0.5252	.1371	-0.0405	-0.0592	-0.0201
-7	-0.0086					-7	-0.3208	.0935	-0.0872	-0.0334	-0.0135
-5	.1264					-5	-0.1387	.0768	-0.1268	-0.106	-0.0093
-3	.2699					-3	.1189	.0706	-0.1825	.0195	-0.0078
-2	.3556					-2	.2676	.0746	-0.2072	.0363	-0.0084
-1	.4006					-1	.3828	.0819	-0.2208	.0500	-0.0092
0	.4627					0	.4905	.0935	-0.2263	.0623	-0.0104
1	.5377					1	.5834	.1073	-0.2307	.0496	-0.0118
2	.5869					2	.6788	.1242	-0.2389	.0852	-0.0135
3	.6341					3	.7754	.1453	-0.2455	.0953	-0.0154
5	.7476					5	.9315	.1919	-0.2631	.1121	-0.0202
7	.8483					7	1.0628	.2541	-0.2828	.1337	-0.0271
10	.8804					10	1.2338	.3497	-0.3047	.1520	-0.0383
15	.8547					15	1.4022	.4837	-0.2992	.1664	-0.0565
20	.8054										
25	.7840										
$M = 0.60$											
$M = 1.00$											
-10	-0.2458	.0632	-0.0718	-0.0290	-0.0082	-10	-0.4975	.1319	-0.0383	-0.0563	-0.0191
-7	.0025	.0226	-0.1165	.0019	-0.0017	-7	-0.3016	.0911	-0.0830	-0.0314	-0.0130
-5	.1679	.0189	-0.1175	.0213	-0.0017	-5	-0.1354	.0765	-0.1190	-0.108	-0.0095
-3	.3358	.0248	-0.1204	.0403	-0.0015	-3	.0724	.0712	-0.1618	.0141	-0.0082
-2	.4159	.0292	-0.1146	.0507	-0.0018	-2	.1971	.0730	-0.1828	.0279	-0.0080
-1	.4872	.0362	-0.1204	.0595	-0.0021	-1	.3218	.0791	-0.2038	.0424	-0.0085
0	.5509	.0432	-0.1155	.0663	-0.0025	0	.4465	.0885	-0.2170	.0556	-0.0094
1	.6211	.0525	-0.1146	.0748	-0.0033	1	.5474	.1016	-0.2238	.0685	-0.0108
2	.6935	.0621	-0.1068	.0831	-0.0047	2	.6317	.1174	-0.2301	.0786	-0.0127
3	.7571	.0751	-0.1082	.0912	-0.0059	3	.7148	.1357	-0.2353	.0885	-0.0146
5	.8779	.1058	-0.1038	.1059	-0.0095	5	.8573	.1793	-0.2501	.1040	-0.0191
7	.9503	.1473	-0.1082	.1189	-0.0168	7	1.0045	.2365	-0.2711	.1250	-0.0257
10	.9656	.2278	-0.1520	.1240	-0.0129	10	1.1613	.3352	-0.3005	.1448	-0.0368
15	.9020	.3226	-0.1762	.1198	-0.0196						
20	.9020	.4198	-0.1913	.1176	-0.0130						
25	.8800	.5094	-0.1937	.1134	-0.0158						
$M = 0.80$											
$M = 1.05$											
-10	-0.2935	.0774	-0.0970	-0.0331	-0.0106	-10	-0.4827	.1271	-0.0342	-0.0550	-0.0183
-7	-0.0964	.0431	-0.1451	-0.0085	-0.0043	-7	-0.2892	.0882	-0.0776	-0.0303	-0.0125
-5	.0875	.0333	-0.1523	.0126	-0.0033	-5	-0.1298	.0733	-0.1116	-0.108	-0.0094
-3	.2656	.0387	-0.1504	.0329	-0.0036	-3	.0524	.0695	-0.1501	.0113	-0.0082
-2	.3465	.0449	-0.1513	.0430	-0.0039	-2	.1548	.0706	-0.1662	.0234	-0.0085
-1	.4355	.0531	-0.1530	.0531	-0.0048	-1	.2687	.0767	-0.1854	.0367	-0.0090
0	.5164	.0615	-0.1507	.0629	-0.0055	0	.3939	.0846	-0.2030	.0504	-0.0093
1	.6047	.0702	-0.1491	.0751	-0.0068	1	.4964	.0965	-0.2130	.0632	-0.0107
2	.6754	.0778	-0.1328	.0826	-0.0077	2	.5829	.1111	-0.2181	.0732	-0.0125
3	.8122	.0959	-0.1393	.0989	-0.0095	3	.6740	.1299	-0.2267	.0836	-0.0143
5	.9211	.1299	-0.1425	.1151	-0.0137	5	.8106	.1719	-0.2433	.0995	-0.0186
7	1.0182	.1791	-0.1497	.1289	-0.0198	7	.9541	.2279	-0.2599	.1198	-0.0252
10	1.0917	.1274	-0.1572	.1423	-0.0292	10	1.1066	.3214	-0.2901	.1388	-0.0354
15	.9593	.1689	-0.1940	.1234	-0.0411						
20	.9446	.4444	-0.2038	.1211	-0.0534						
25	.9299	.5456	-0.2168	.1194	-0.0653						
$M = 0.90$											
$M = 1.10$											
-10	-0.4934	.1200	-0.0642	-0.0589	-0.0176	-10	-0.4700	.1209	-0.0310	-0.0534	-0.0178
-7	-0.3216	.0775	-0.0901	-0.0346	-0.0108	-7	-0.2859	.0854	-0.0727	-0.0296	-0.0120
-5	-0.0924	.0592	-0.1376	-0.0071	-0.0066	-5	-0.1391	.0722	-0.1049	-0.1119	-0.0092
-3	.1810	.0599	-0.1860	.0241	-0.0058	-3	.0449	.0670	-0.1405	.0109	-0.0082
-2	.3033	.0647	-0.1918	.0380	-0.0067	-2	.1490	.0692	-0.1599	.0221	-0.0083
-1	.4075	.0742	-0.2004	.0498	-0.0079	-1	.2585	.0746	-0.1759	.0343	-0.0084
0	.4960	.0838	-0.2033	.0622	-0.0087	0	.3571	.0824	-0.1919	.0460	-0.0094
1	.5637	.0871	-0.1889	.0698	-0.0093	1	.4776	.0940	-0.2065	.0604	-0.0110
2	.6666	.1047	-0.1918	.0824	-0.0112	2	.5587	.1075	-0.2137	.0700	-0.0124
3	.7577	.1239	-0.2033	.0933	-0.0130	3	.6485	.1258	-0.2210	.0800	-0.0143
5	.8931	.1623	-0.2033	.1097	-0.0176	5	.7866	.1665	-0.2341	.0955	-0.0183
7	1.0181	.2222	-0.2292	.1294	-0.0239	7	.9246	.2219	-0.3451	.1153	-0.0249
10	1.1639	.3067	-0.2379	.1471	-0.0348	10	1.0648	.3124	-0.2743	.1323	-0.0351
15	1.0311	.3803	-0.2223	.1299	-0.0442						

TABLE 6.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 6 MODEL

 $\frac{b}{c} = 0.06$        $\frac{C_f}{c} = \text{NONE}$ 

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
$M = 0.40$											
-10	-0.6162					-10	-0.8915	0.1751	0.1455	-0.1119	-0.0264
-7	-0.4843					-7	-0.6596	0.1005	0.1016	-0.0829	-0.0149
-5	-0.3542					-5	-0.4932	0.0643	0.0676	-0.0617	-0.0096
-3	-0.2169					-3	-0.3056	0.0419	0.0373	-0.0385	-0.0056
-2	-0.1500					-2	-0.1897	0.0332	0.0233	-0.0239	-0.0035
-1	-0.0813					-1	-0.0822	0.0280	0.0093	-0.0119	-0.0023
0	-0.0108					0	-0.0211	0.0290	0.0000	0.0014	-0.0020
1	0.0813					1	0.1286	0.0332	-0.0131	0.0136	-0.0015
2	0.1536					2	0.2297	0.0399	-0.0233	0.0252	-0.0016
3	0.2440					3	0.3309	0.0513	-0.0396	0.0375	-0.0024
5	0.3650					5	0.4995	0.0798	-0.0699	0.0580	-0.0050
7	0.5096					7	0.7060	0.1280	-0.1165	0.0818	-0.0100
10	0.6813					10	0.9463	0.2145	-0.1818	0.1108	-0.0199
15	0.8187										
20	1.0012										
25	0.7301										
$M = 0.60$											
$M = 0.60$											
-10	-0.7031	0.1239	0.0610	-0.0901	-0.0177	-10	-0.8383	0.1659	0.1564	-0.1046	-0.0251
-7	-0.5906	0.0585	0.0108	-0.0722	-0.0086	-7	-0.6181	0.0989	0.0929	-0.0771	-0.0146
-5	-0.4312	0.0272	-0.0008	-0.0522	-0.0039	-5	-0.4585	0.0640	0.0648	-0.0579	-0.0096
-3	-0.2681	0.0111	-0.0208	-0.0322	-0.0015	-3	-0.2787	0.0427	0.0402	-0.0360	-0.0056
-2	-0.1744	0.0073	-0.0112	-0.0218	-0.0004	-2	-0.1838	0.0368	0.0268	-0.0239	-0.0045
-1	-0.0937	0.0051	-0.0033	-0.0115	-0.0001	-1	-0.0869	0.0342	0.0121	-0.0124	-0.0031
0	-0.0056	0.0032	0.0000	-0.0006	0.0003	0	0.0182	0.0333	0.0000	0.0007	-0.0027
1	0.0787	0.0064	0.0008	0.0091	0.0005	1	0.1192	0.0378	-0.0134	0.0118	-0.0027
2	0.1669	0.0101	0.0021	0.0188	0.0006	2	0.2161	0.0442	-0.0290	0.0235	-0.0027
3	0.2587	0.0171	0.0021	0.0291	0.0001	3	0.3131	0.0551	-0.0425	0.0346	-0.0035
5	0.4219	0.0088	0.0021	0.0479	-0.0002	5	0.4787	0.0815	-0.0715	0.0546	-0.0059
7	0.5981	0.0673	0.0041	0.0677	-0.0028	7	0.6403	0.1222	-0.1028	0.0742	-0.0098
10	0.7537	0.1378	-0.0436	0.0880	-0.0102	10	0.8726	0.1992	-0.1622	0.1013	-0.0185
15	0.7856	0.2417	-0.1000	0.0995	-0.0238						
20	0.7725	0.3217	-0.1120	0.0983	-0.0349						
25	0.8044	0.4177	-0.1203	0.0998	-0.0470						
$M = 0.80$											
$M = 0.80$											
-10	-0.8201	0.1350	0.0567	-0.1060	-0.0201	-10	-0.7909	0.1583	0.1449	-0.0988	-0.0239
-7	-0.7026	0.0765	0.0346	-0.0882	-0.0114	-7	-0.5874	0.0934	0.0901	-0.0737	-0.0144
-5	-0.5576	0.0381	0.0139	-0.0684	-0.0059	-5	-0.4459	0.0619	0.0652	-0.0558	-0.0098
-3	-0.3326	0.0120	-0.0083	-0.0405	-0.0019	-3	-0.2695	0.0405	0.0395	-0.0345	-0.0056
-2	-0.2075	0.0068	-0.0061	-0.0255	-0.0010	-2	-0.1745	0.0353	0.0249	-0.0229	-0.0044
-1	-0.1100	0.0049	-0.0025	-0.0142	-0.0004	-1	-0.0756	0.0333	0.0107	-0.0113	-0.0029
0	-0.0075	0.0059	0.0000	-0.0020	0.0001	0	0.0174	0.0338	-0.0004	0.0000	-0.0028
1	0.1100	0.0089	0.0011	0.0121	0.0001	1	0.1144	0.0386	-0.0163	0.0110	-0.0026
2	0.2150	0.0129	0.0061	0.0235	0.0001	2	0.2074	0.0448	-0.0292	0.0220	-0.0028
3	0.3101	0.0196	0.0144	0.0344	-0.0002	3	0.2947	0.0539	-0.0407	0.0323	-0.0034
5	0.5101	0.0451	0.0028	0.0587	-0.0018	5	0.4536	0.0791	-0.0665	0.0511	-0.0056
7	0.6926	0.0889	-0.0199	0.0817	-0.0062	7	0.6145	0.1196	-0.0986	0.0706	-0.0095
10	0.8251	0.1534	-0.0429	0.1003	-0.0140	10	0.8200	0.1916	-0.1509	0.0941	-0.0171
15	0.8301	0.2484	-0.1126	0.0991	-0.0265						
20	0.8677	0.3554	-0.1372	0.1072	-0.0397						
25	0.8602	0.4415	-0.1493	0.1068	-0.0504						
$M = 0.90$											
$M = 0.90$											
-10	-0.8895	0.1661	0.1416	-0.1107	-0.0246	-10	-0.7660	0.1544	0.1402	-0.0959	-0.0233
-7	-0.6908	0.0971	0.0976	-0.0850	-0.0139	-7	-0.5741	0.0912	0.0928	-0.0709	-0.0142
-5	-0.5143	0.0564	0.0512	-0.0625	-0.0078	-5	-0.4249	0.0614	0.0639	-0.0534	-0.0127
-3	-0.3046	0.0283	0.0147	-0.0368	-0.0031	-3	-0.2572	0.0399	0.0371	-0.0326	-0.0058
-2	-0.2008	0.0212	0.0073	-0.0239	-0.0020	-2	-0.1640	0.0349	0.0235	-0.0217	-0.0042
-1	-0.1170	0.0158	0.0098	-0.0136	-0.0010	-1	-0.0708	0.0339	0.0124	-0.0112	-0.0030
0	-0.0066	0.0141	0.0039	0.0000	-0.0002	0	0.0168	0.0339	-0.0008	0.0000	-0.0026
1	0.0993	0.0184	-0.0024	0.0125	-0.0001	1	0.1118	0.0376	-0.0157	0.0109	-0.0024
2	0.1920	0.0238	-0.0024	0.0243	-0.0003	2	0.2032	0.0440	-0.0280	0.0217	-0.0026
3	0.2980	0.0348	-0.0147	0.0393	-0.0009	3	0.2870	0.0531	-0.0412	0.0317	-0.0033
5	0.4679	0.0608	-0.0391	0.0596	-0.0032	5	0.4455	0.0788	-0.0680	0.0498	-0.0054
7	0.6555	0.1064	-0.0781	0.0821	-0.0080	7	0.6001	0.1178	-0.0989	0.0688	-0.0091
10	0.8586	0.1786	-0.1099	0.1057	-0.0172	10	0.8052	0.1866	-0.1402	0.0920	-0.0173
15	1.1345	0.3289	-0.1587	0.1336	-0.0349						
20	0.9800	0.3897	-0.1782	0.1196	-0.0212						
$M = 1.10$											
$M = 1.10$											
-10	-0.7660	0.1544	0.1402	-0.0959	-0.0233						
-7	-0.5741	0.0912	0.0928	-0.0709	-0.0142						
-5	-0.4249	0.0614	0.0639	-0.0534	-0.0127						
-3	-0.2572	0.0399	0.0371	-0.0326	-0.0058						
-2	-0.1640	0.0349	0.0235	-0.0217	-0.0042						
-1	-0.0708	0.0339	0.0124	-0.0112	-0.0030						
0	0.0168	0.0339	-0.0008	0.0000	-0.0026						
1	0.1118	0.0376	-0.0157	0.0109	-0.0024						
2	0.2032	0.0440	-0.0280	0.0217	-0.0026						
3	0.2870	0.0531	-0.0412	0.0317	-0.0033						
5	0.4455	0.0788	-0.0680	0.0498	-0.0054						
7	0.6001	0.1178	-0.0989	0.0688	-0.0091						
10	0.8052	0.1866	-0.1402	0.0920	-0.0173						

TABLE 6.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 6 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.20$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$						$M = 0.95$					
-10	+.3355					-10	-.7749	.1613	.0551	-.0924	-.0246
-7	-.1435					-7	-.5520	.0973	.0125	-.0651	-.0149
-5	.0072					-5	-.3791	.0692	-.0221	-.0425	-.0105
-3	.1507					-3	-.1979	.0461	-.0544	-.0199	-.0057
-2	.2260					-2	-.0771	.0435	-.0739	-.0064	-.0043
-1	.2960					-1	.0583	.0205	-.1074	.0098	-.0038
0	.3570					0	.2354	.0553	-.1525	.0290	-.0043
1	.4180					1	.3979	.0676	-.1908	.0482	-.0055
2	.4790					2	.5270	.0820	-.2087	.0634	-.0065
3	.5436					3	.6436	.0999	-.2216	.0772	-.0085
5	.6605					5	.8353	.1434	-.2433	.0981	-.0134
7	.7858					7	.9998	.2038	-.2728	.1190	-.0197
10	.8575										
15	1.0513										
20	1.2127										
25	1.2917										
$M = 0.60$						$M = 1.00$					
-10	-.3858	.0800	-.0592	-.0462	-.0114	-10	-.7322	.1560	.0561	-.0885	-.0236
-7	-.1692	.0257	-.1258	-.0179	-.0028	-7	-.5127	.0967	.0064	-.0600	-.0147
-5	-.0130	.0132	-.1312	.0012	-.0012	-5	-.3431	.0711	-.0304	-.0391	-.0104
-3	.1376	.0141	-.1258	.0182	-.0008	-3	-.1536	.0565	-.0664	-.0158	-.0074
-2	.2203	.0169	-.1185	.0269	-.0007	-2	-.0479	.0555	-.0856	-.0036	-.0059
-1	.2956	.0206	-.1180	.0360	-.0009	-1	.0559	.0540	-.1061	.0087	-.0053
0	.3849	.0260	-.1177	.0463	-.0011	0	.1855	.0589	-.1293	.0236	-.0051
1	.5132	.0335	-.1168	.0608	-.0015	1	.3152	.0662	-.1571	.0391	-.0058
2	.5783	.0420	-.1168	.0686	-.0018	2	.4648	.0805	-.1924	.0568	-.0070
3	.6582	.0536	-.1135	.0779	-.0018	3	.6005	.0982	-.2127	.0713	-.0084
5	.7977	.0824	-.1065	.0945	-.0039	5	.7601	.1374	-.2365	.0907	-.0122
7	.9130	.1303	-.1135	.1092	-.0089	7	.9437	.1938	-.2613	.1114	-.0186
10	.9818	.2103	-.1567	.1207	-.0192						
15	.9037	.3046	-.1855	.1176	-.0324						
$M = 0.80$						$M = 1.05$					
-10	-.4676	.0955	-.0620	-.0544	-.0147	-10	-.7116	.1496	.0548	-.0839	-.0224
-7	-.4056	.0482	-.0922	-.0460	-.0082	-7	-.4858	.0941	.0072	-.0570	-.0143
-5	-.1947	.0234	-.1237	-.0209	-.0034	-5	-.3233	.0682	-.0277	-.0368	-.0105
-3	.0459	.0171	-.1278	.0072	-.0015	-3	-.1473	.0550	-.0626	-.0152	-.0078
-2	.1625	.0198	-.1284	.0203	-.0013	-2	-.0516	.0541	-.0806	-.0043	-.0056
-1	.2704	.0257	-.1306	.0323	-.0012	-1	.0497	.0532	-.0997	.0080	-.0054
0	.3907	.0341	-.1377	.0460	-.0012	0	.1626	.0564	-.1189	.0204	-.0055
1	.5122	.0448	-.1432	.0604	-.0016	1	.2831	.0645	-.1430	.0350	-.0059
2	.6697	.0579	-.1526	.0775	-.0024	2	.3940	.0753	-.1654	.0483	-.0065
3	.8012	.0747	-.1542	.0925	-.0036	3	.5318	.0696	-.1993	.0638	-.0079
5	.9376	.1108	-.1542	.1128	-.0075	5	.7230	.1307	-.2251	.0860	-.0119
7	1.0294	.1616	-.1596	.1268	-.0127	7	.8837	.1834	-.2505	.1049	-.0176
10	1.1286	.2379	-.1652	.1393	-.0224						
15	.9376	.3171	-.1997	.1168	-.0346						
$M = 0.90$						$M = 1.10$					
-10	-.6788	.1362	-.0003	-.0822	-.0211	-10	-.6749	.1452	.0547	-.0815	-.0218
-7	-.5364	.0819	-.0075	-.0636	-.0130	-7	-.4726	.0905	.0069	-.0548	-.0139
-5	-.3810	.0485	-.0305	-.0423	-.0077	-5	-.3163	.0217	-.0242	-.0363	-.0102
-3	-.1686	.0328	-.0668	-.0167	-.0042	-3	-.1361	.0543	-.0582	-.0146	-.0074
-2	.0175	.0312	-.1216	.0046	-.0023	-2	-.0405	.0533	-.0795	-.0033	-.0054
-1	.1883	.0350	-.1506	.0241	-.0022	-1	.0570	.0520	-.0958	.0077	-.0057
0	.3131	.0430	-.1589	.0383	-.0026	0	.1582	.0543	-.1143	.0196	-.0056
1	.4270	.0539	-.1651	.0516	-.0038	1	.2722	.0615	-.1347	.0330	-.0062
2	.5386	.0668	-.1676	.0636	-.0050	2	.3788	.0941	-.1562	.0455	-.0073
3	.6525	.0862	-.1869	.0764	-.0066	3	.4984	.0868	-.1814	.0592	-.0082
5	.8167	.1238	-.2039	.0985	-.0106	5	.6860	.1221	-.2148	.0812	-.0120
7	.9765	.1777	-.2184	.1180	-.0166	7	.8515	.1741	-.2408	.1003	-.0176
10	1.1911	.2746	-.2625	.1396	-.0277						

TABLE 6.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 6 MODEL - Continued

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.30$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$											
-10	-0.2418					-10	-0.6551	0.1554	-0.0069	-0.0712	-0.0218
-7	-0.0412					-7	-0.4409	0.1028	-0.0515	-0.0443	-0.0135
-5	0.1021					-5	-0.2599	0.0782	-0.0837	-0.0215	-0.0096
-3	0.2507					-3	-0.0374	0.0645	-0.1251	0.0062	-0.0056
-2	0.3349					-2	0.1248	0.0685	-0.1573	0.0239	-0.0054
-1	0.4083					-1	0.2911	0.0773	-0.1950	0.0432	-0.0056
0	0.4567					0	0.4471	0.0890	-0.2171	0.0592	-0.0062
1	0.5176					1	0.5677	0.1043	-0.2203	0.0727	-0.0072
2	0.5821					2	0.6696	0.1248	-0.2291	0.0427	-0.0088
3	0.6358					3	0.7694	0.1463	-0.2410	0.0488	-0.0109
5	0.7522					5	0.9358	0.1923	-0.2548	0.0584	-0.0159
7	0.8686					7	1.1146	0.2562	-0.2778	0.0676	-0.0227
10	0.9134										
15	1.0925										
20	1.2358										
25	1.2967										
$M = 0.60$											
-10	-0.3063	0.0835	-0.0673	-0.0401	-0.0105	-10	-0.6374	0.1519	-0.0066	-0.0682	-0.0215
-7	-0.0696	0.0319	-0.0665	0.0149	-0.0032	-7	-0.4123	0.1009	-0.0494	-0.0408	-0.0140
-5	0.1095	0.0206	-0.1302	0.0364	-0.0027	-5	-0.2390	0.0813	-0.0837	-0.0190	-0.0100
-3	0.2952	0.0251	-0.1302	0.0362	-0.0022	-3	-0.0319	0.0740	-0.1229	0.0060	-0.0068
-2	0.3898	0.0301	-0.1322	0.0469	-0.0022	-2	0.0837	0.0725	-0.1419	0.0197	-0.0065
-1	0.4734	0.0369	-0.1219	0.0559	-0.0025	-1	0.2151	0.0764	-0.1661	0.0350	-0.0067
0	0.5532	0.0438	-0.1199	0.0656	-0.0028	0	0.3884	0.0862	-0.2057	0.0548	-0.0069
1	0.6126	0.0507	-0.1138	0.0721	-0.0029	1	0.5218	0.1029	-0.2177	0.0672	-0.0076
2	0.6869	0.1077	-0.1035	0.0817	-0.0030	2	0.6174	0.1195	-0.2238	0.0785	-0.0086
3	0.7704	0.0780	-0.1035	0.0901	-0.0033	3	0.7230	0.1401	-0.2291	0.0919	-0.0108
5	0.9096	0.1123	-0.0945	0.1069	-0.0064	5	0.8803	0.1841	-0.2485	0.1096	-0.0150
7	0.9653	0.1639	-0.1138	0.1163	-0.0122	7	1.0476	0.2439	-0.2688	0.1270	-0.0216
10	1.0581	0.2519	-0.1454	0.1292	-0.0230						
15	1.0136	0.3616	-0.1753	0.1295	-0.0383						
20	0.9245	0.4327	-0.1733	0.1163	-0.0484						
$M = 0.80$											
-10	-0.3654	0.0931	-0.0992	-0.0449	-0.0115	-10	-0.6111	0.1474	-0.0021	-0.0657	-0.0208
-7	-0.2291	0.0476	-0.1408	0.0259	-0.0049	-7	-0.4010	0.0967	-0.0452	-0.0391	-0.0134
-5	-0.0248	0.0292	-0.1518	0.0020	-0.0024	-5	-0.2349	0.0766	-0.0790	-0.0182	-0.0101
-3	0.1858	0.0338	-0.1436	0.0222	-0.0022	-3	-0.0382	0.0714	-0.1149	0.0046	-0.0067
-2	0.2948	0.0393	-0.1397	0.0338	-0.0022	-2	0.0726	0.0700	-0.1347	0.0181	-0.0068
-1	0.3864	0.0469	-0.1381	0.0441	-0.0023	-1	0.1814	-0.0198	-0.1517	0.0181	-0.0068
0	0.4954	0.0566	-0.1343	0.0565	-0.0025	0	0.3132	0.0835	-0.1761	0.0451	-0.0071
1	0.6131	0.0704	-0.1397	0.0696	-0.0028	1	0.4583	0.0982	-0.2015	0.0606	-0.0078
2	0.7679	0.0871	-0.1408	0.0866	-0.0038	2	0.5767	0.1146	-0.2154	0.0732	-0.0087
3	0.8769	0.1063	-0.1436	0.1022	-0.0052	3	0.6722	0.1352	-0.2230	0.0856	-0.0104
5	1.0058	0.1480	-0.1452	0.1191	-0.0093	5	0.8250	0.1756	-0.2357	0.1026	-0.0144
7	1.0949	0.1985	-0.1381	0.1311	-0.0156	7	0.9930	0.2353	-0.2577	0.1211	-0.0208
10	1.1147	0.2687	-0.1518	0.1431	-0.0255						
15	0.9686	0.3545	-0.1874	0.1291	-0.0369						
$M = 0.90$											
-10	-0.5904	0.1349	-0.0517	-0.0651	-0.0195	-10	-0.5838	0.1427	-0.0032	-0.0324	-0.0203
-7	-0.4548	0.0812	-0.0517	-0.0448	-0.0117	-7	-0.3874	0.0944	-0.0422	-0.0193	-0.0135
-5	-0.2690	0.0559	-0.0856	-0.0209	-0.0070	-5	-0.2240	0.0749	-0.0739	-0.0089	-0.0096
-3	0.0437	0.0462	-0.1509	0.0154	-0.0040	-3	-0.0367	0.0690	-0.1105	0.0025	-0.0073
-2	0.2034	0.0516	-0.1678	0.0329	-0.0032	-2	0.0734	0.0682	-0.1287	0.0085	-0.0065
-1	0.3324	0.0623	-0.1799	0.0472	-0.0039	-1	0.1652	0.0722	-0.1430	0.0144	-0.0068
0	0.4461	0.0715	-0.1727	0.0596	-0.0044	0	0.2846	0.0803	-0.1633	0.0208	-0.0070
1	0.5467	0.0892	-0.1872	0.0720	-0.0053	1	0.4223	0.0948	-0.1876	0.0285	-0.0076
2	0.6451	0.1027	-0.1799	0.0823	-0.0064	2	0.5471	0.1111	-0.2071	0.0357	-0.0087
3	0.7544	0.1253	-0.1920	0.0962	-0.0083	3	0.6463	0.1300	-0.2153	0.0412	-0.0105
5	0.9118	0.1726	-0.2114	0.1157	-0.0131	5	0.7987	0.1697	-0.2291	0.0496	-0.0143
7	1.0715	0.2291	-0.2235	0.1334	-0.0154						
$M = 1.00$											
-10	-0.5157	0.1427	-0.0324	-0.0193	-0.0070						
-7	-0.3874	0.0944	-0.0422	-0.0193	-0.0070						
-5	-0.2240	0.0749	-0.0739	-0.0089	-0.0070						
-3	-0.0367	0.0690	-0.1105	0.0025	-0.0073						
-2	0.0734	0.0682	-0.1287	0.0085	-0.0065						
-1	0.1652	0.0722	-0.1430	0.0144	-0.0068						
0	0.2846	0.0803	-0.1633	0.0208	-0.0070						
1	0.4223	0.0948	-0.1876	0.0285	-0.0076						
2	0.5471	0.1111	-0.2071	0.0357	-0.0087						
3	0.6463	0.1300	-0.2153	0.0412	-0.0105						
5	0.7987	0.1697	-0.2291	0.0496	-0.0143						

TABLE 6.- THE AERODYNAMIC CHARACTERISTICS OF THE ASPECT RATIO 6 MODEL - Concluded

$$\frac{t}{c} = 0.06 \quad \frac{c_f}{c} = 0.40$$

$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$	$\alpha$ , deg	$C_L$	$C_D$	$C_M$	$C_l$	$C_n$
$M = 0.40$						$M = 0.95$					
-10	-0.2076					-10	-0.5620	0.1454	-0.0497	-0.0694	-0.0408
-7	0.0055					-7	-0.3487	0.3880	-0.0939	-0.0413	-0.0131
-5	0.1694					-5	-0.1603	0.3245	-0.1345	-0.0177	-0.0093
-3	0.3151					-3	0.1072	0.2959	-0.1929	0.0147	-0.0074
-2	0.3879					-2	0.2758	0.3194	-0.2192	0.0337	-0.0073
-1	0.4571					-1	0.4038	0.3583	-0.2431	0.0480	-0.0077
0	0.5154					0	0.5131	0.4115	-0.2431	0.0606	-0.0086
1	0.5973					1	0.6266	0.4761	-0.2477	0.0733	-0.0098
2	0.6520					2	0.7306	0.4393	-0.2569	0.0859	-0.0115
3	0.7066					3	0.8285	0.4597	-0.2615	0.0943	-0.0137
5	0.8341					5	0.9784	0.2084	-0.2836	0.1159	-0.0190
7	0.9324					7	1.1574	0.2784	-0.3030	0.1331	-0.0264
10	0.9215										
15	1.1145										
20	1.2311										
25	1.2639										
$M = 0.60$						$M = 1.00$					
-10	-0.2495	0.0659	-0.0748	~0.0318	-0.0071	-10	-0.5263	0.1383	-0.0459	-0.0645	-0.0197
-7	0.0170	0.0185	-0.1207	0.0028	-0.0022	-7	-0.3170	0.0990	-0.0944	-0.0379	-0.0132
-5	0.2002	0.0194	-0.1240	0.0238	-0.0014	-5	-0.1535	0.0843	-0.1297	-0.0177	-0.0098
-3	0.3824	0.0255	-0.1270	0.0443	-0.0010	-3	0.0558	0.0785	-0.1729	0.0086	-0.0087
-2	0.4683	0.0321	-0.1258	0.0550	-0.0012	-2	0.2014	0.0818	-0.2002	0.0255	-0.0080
-1	0.5534	0.0391	-0.1136	0.0636	-0.0017	-1	0.3609	0.0892	-0.2329	0.0431	-0.0082
0	0.6193	0.0474	-0.1165	0.0718	-0.0019	0	0.4885	0.1025	-0.2355	0.0573	-0.0088
1	0.6987	0.0604	-0.1165	0.0810	-0.0020	1	0.5941	0.1186	-0.2417	0.0690	-0.0098
2	0.7799	0.0729	-0.1082	0.0895	-0.0024	2	0.6978	0.1358	-0.2461	0.0816	-0.0117
3	0.8554	0.0906	-0.1090	0.0978	-0.0032	3	0.7835	0.1569	-0.2567	0.0919	-0.0135
5	0.9744	0.1254	-0.1061	0.1121	-0.0064	5	0.9291	0.2010	-0.2708	0.1090	-0.0185
7	1.0046	0.1792	-0.1278	0.1173	-0.0130	7	1.1005	0.2667	-0.2946	0.1252	-0.0253
10	1.1216	0.2880	-0.1666	0.1338	-0.0264						
15	0.9366	0.3437	-0.1750	0.1176	-0.0370						
20	0.9177	0.4383	-0.1834	0.1137	-0.0480						
$M = 0.80$						$M = 1.05$					
-10	-0.3272	0.0804	-0.1069	~0.0387	-0.0094	-10	-0.5107	0.1373	-0.0410	-0.0638	-0.0191
-7	-0.1334	0.0443	-0.1436	~0.0143	-0.0042	-7	-0.3156	0.0974	-0.0855	-0.0387	-0.0130
-5	0.0692	0.0295	-0.1525	0.0092	-0.0024	-5	-0.1530	0.0819	-0.1202	-0.0189	-0.0096
-3	0.2618	0.0399	-0.1487	0.0312	-0.0022	-3	0.0593	0.0762	-0.1633	0.0059	-0.0085
-2	0.3524	0.0477	-0.1470	0.0413	-0.0023	-2	0.1683	0.0800	-0.1815	0.0193	-0.0082
-1	0.4430	0.0558	-0.1442	0.0517	-0.0026	-1	0.3060	0.0880	-0.2149	0.0341	-0.0082
0	0.5702	0.0717	-0.1576	0.0662	-0.0033	0	0.4457	0.0988	-0.2259	0.0500	-0.0087
1	0.6825	0.0842	-0.1525	0.0784	-0.0041	1	0.5585	0.1148	-0.2344	0.0619	-0.0097
2	0.8131	0.0982	-0.1459	0.0917	-0.0049	2	0.6523	0.1327	-0.2403	0.0724	-0.0112
3	0.9138	0.1188	-0.1525	0.1055	-0.0066	3	0.7422	0.1524	-0.2479	0.0842	-0.0132
5	1.0271	0.1653	-0.1626	0.1210	-0.0111	5	0.8856	0.1962	-0.2657	0.1006	-0.0179
7	1.1152	0.2198	-0.1682	0.1344	-0.0177	7	1.0482	0.2606	-0.2869	0.1161	-0.0245
10	1.0145	0.2643	-0.1693	0.1251	-0.0262						
15	0.8534	0.3144	-0.1525	0.1222	-0.0388						
20	0.7124	0.3367	-0.1108	0.1214	-0.0529						
$M = 0.90$						$M = 1.10$					
-10	-0.4876	0.1167	-0.0784	~0.0623	-0.0173	-10	-0.4905	0.1319	-0.0211	-0.0632	-0.0184
-7	-0.3258	0.0737	-0.0972	~0.0389	-0.0101	-7	-0.2976	0.0912	-0.0410	-0.0394	-0.0129
-5	-0.1028	0.0581	-0.1432	~0.0117	-0.0059	-5	-0.1470	0.0777	-0.0585	-0.0208	-0.0095
-3	0.1662	0.0559	-0.1756	0.0198	-0.0043	-3	0.0514	0.0741	-0.0789	0.0033	-0.0085
-2	0.2886	0.0623	-0.1794	0.0340	-0.0044	-2	0.1617	0.0777	-0.0878	0.0156	-0.0083
-1	0.3958	0.0721	-0.1809	0.0460	-0.0051	-1	0.2811	0.0845	-0.0984	0.0294	-0.0083
0	0.4986	0.0844	-0.1857	0.0577	-0.0059	0	0.4097	0.0949	-0.1073	0.0439	-0.0087
1	0.6254	0.1054	-0.2051	0.0718	-0.0075	1	0.5236	0.1093	-0.1134	0.0572	-0.0097
2	0.6800	0.1086	-0.1881	0.0796	-0.0076	2	0.6118	0.1260	-0.1162	0.0669	-0.0111
3	0.7828	0.1312	-0.2003	0.0924	-0.0099	3	0.7036	0.1455	-0.1195	0.0782	-0.0132
5	0.9206	0.1742	-0.2099	0.1097	-0.0145	5	0.8414	0.1870	-0.1268	0.0936	-0.0176
7	1.0496	0.2301	-0.2268	0.1253	-0.0214	7	0.9921	0.2457	-0.1561	0.1100	-0.0240
10	1.2376	0.3302	-0.2651	0.1444	-0.0321						

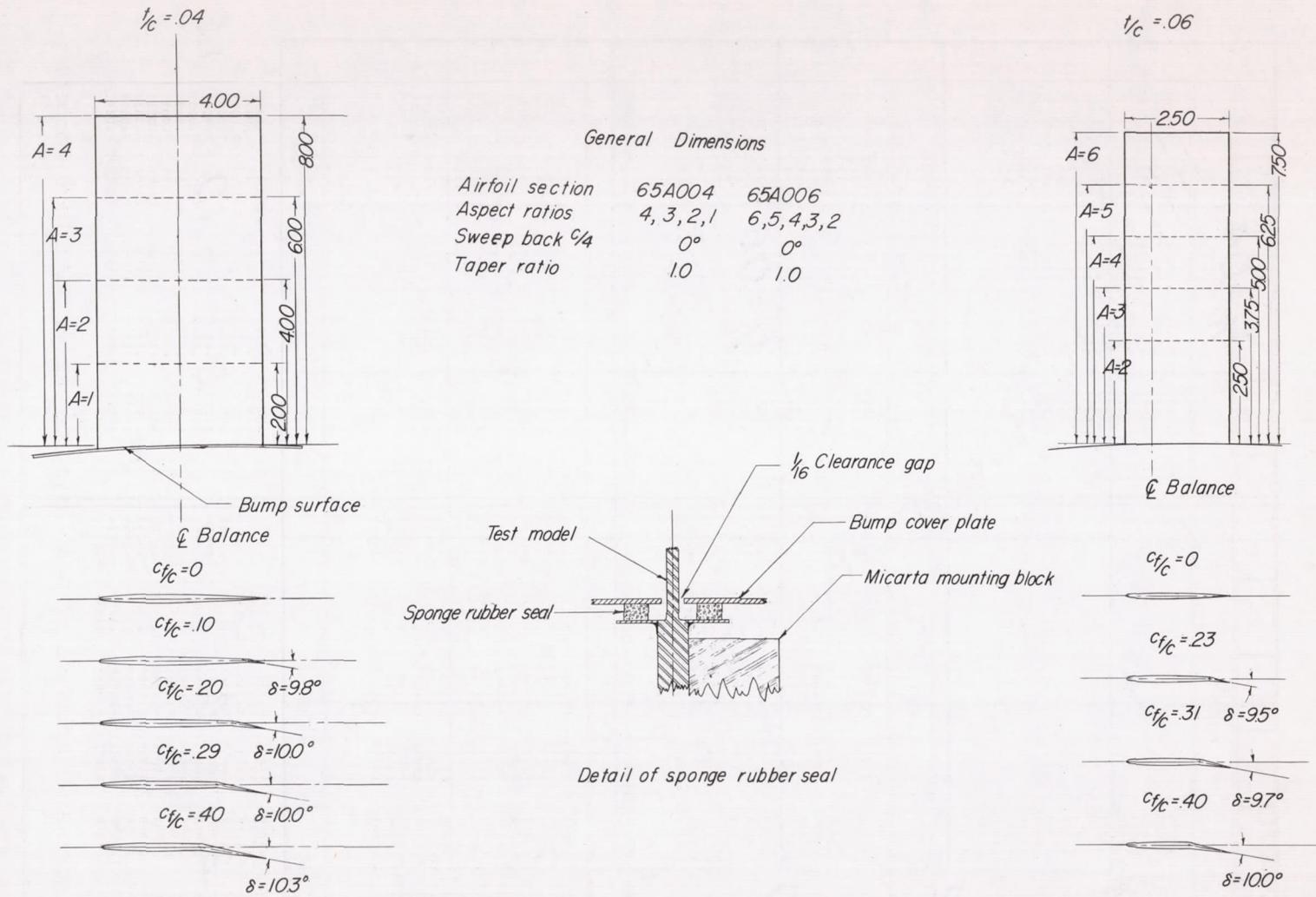


Figure 1.- General dimensions and model geometry. (All dimensions in inches unless otherwise noted.)

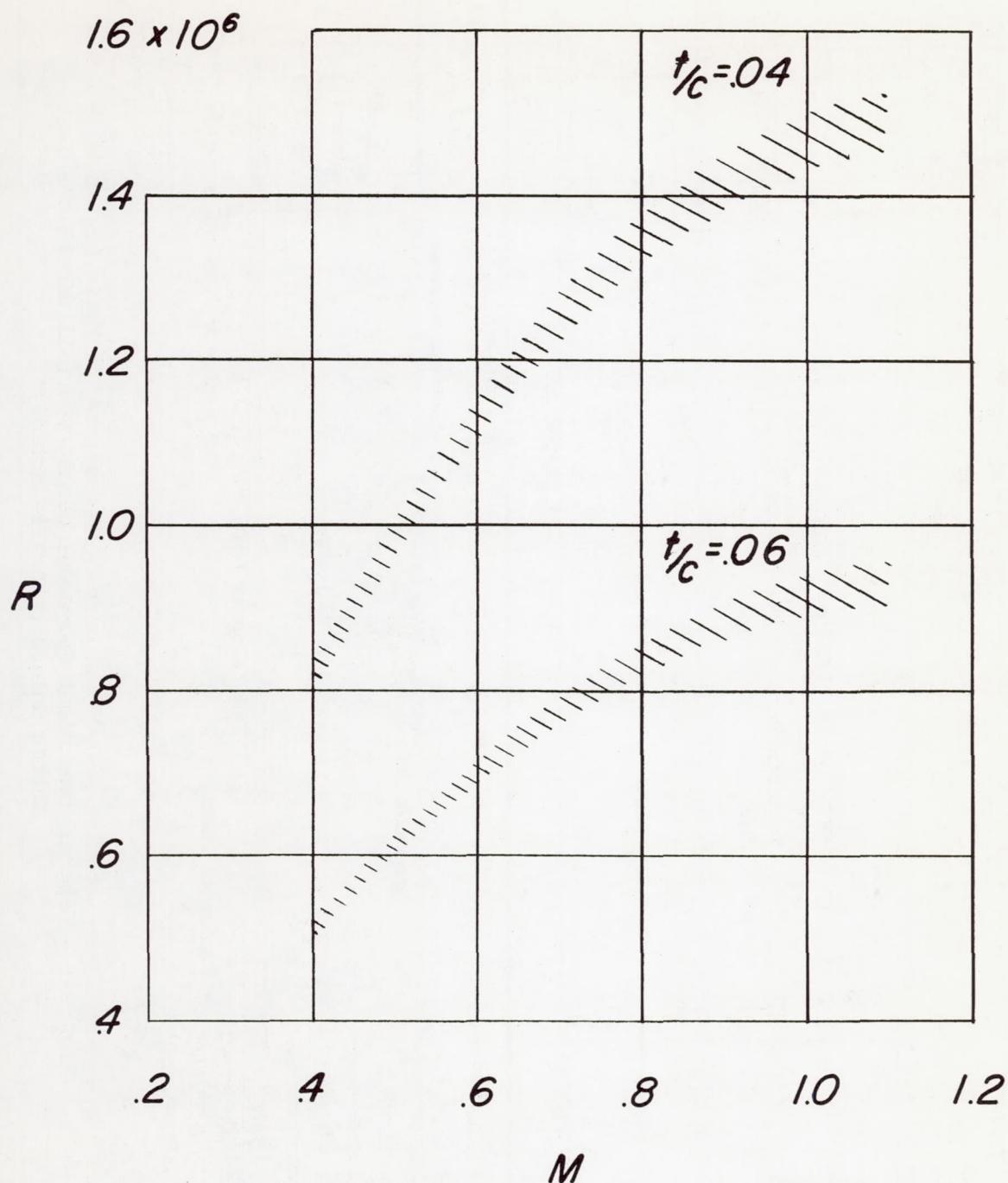
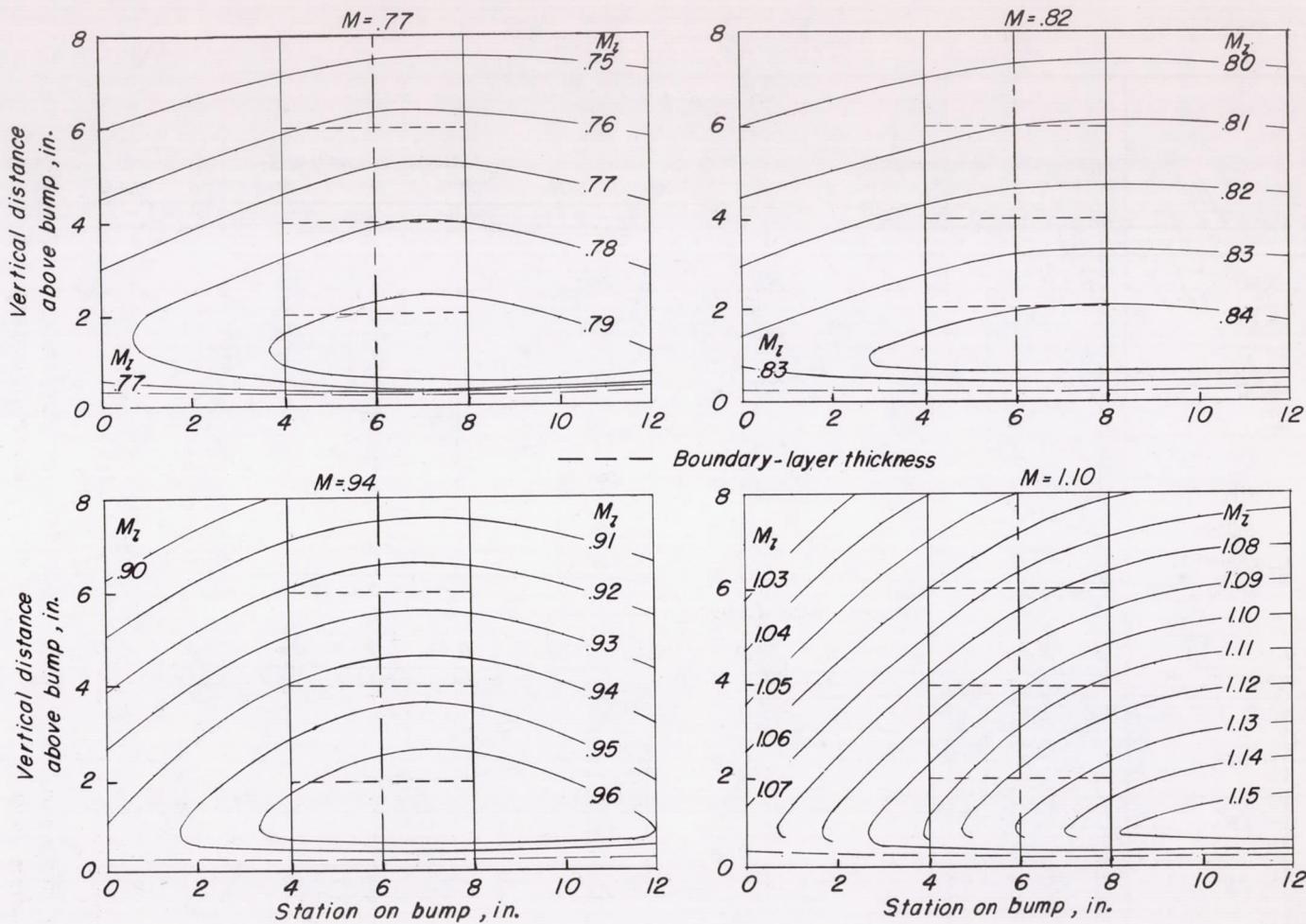
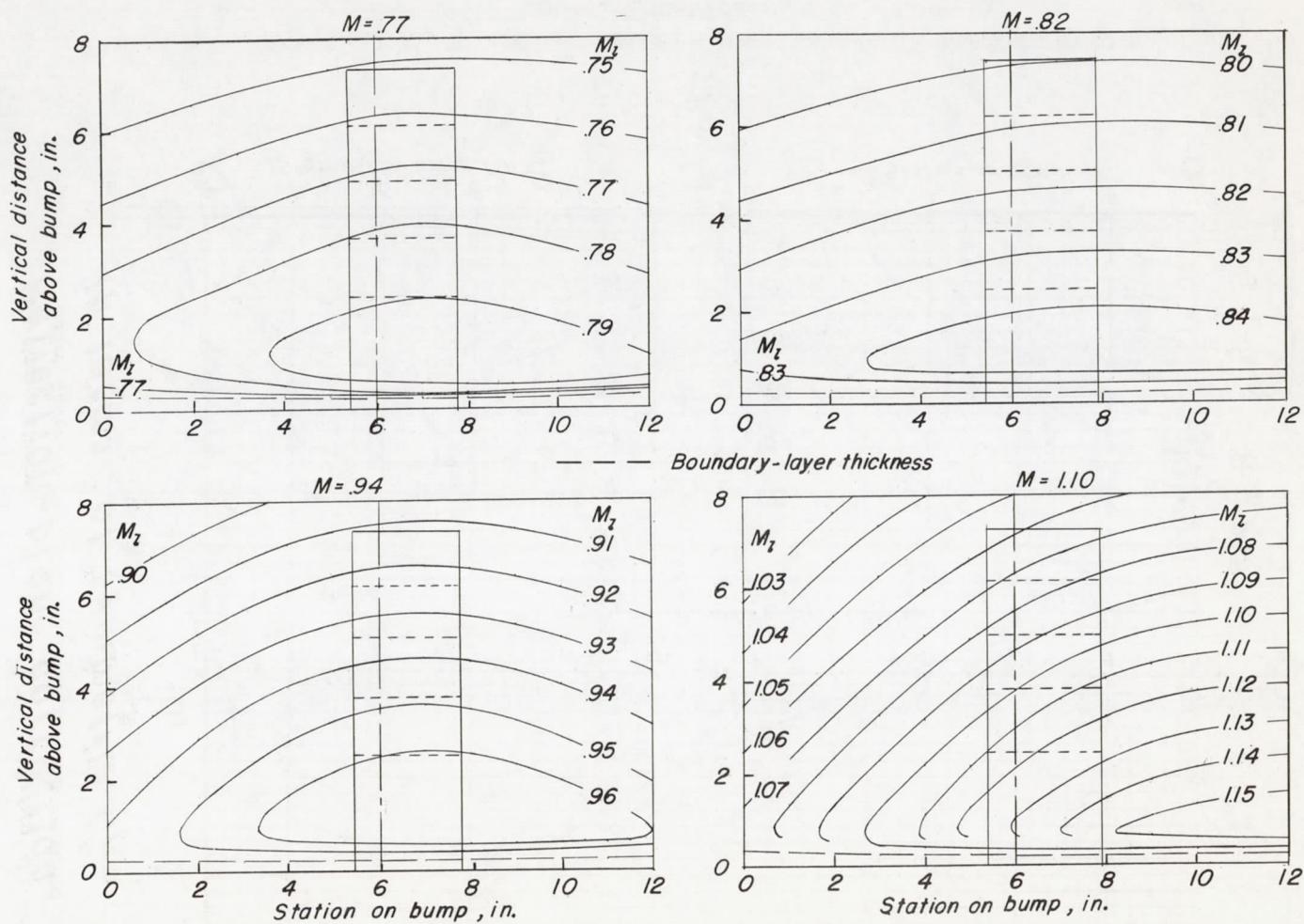


Figure 2.- Variation of test Reynolds number with Mach number.



(a)  $t/c = 0.04$ .

Figure 3.- Typical Mach number contours over transonic bump in region of model location.



(b)  $t/c = 0.06$ .

Figure 3.- Concluded.

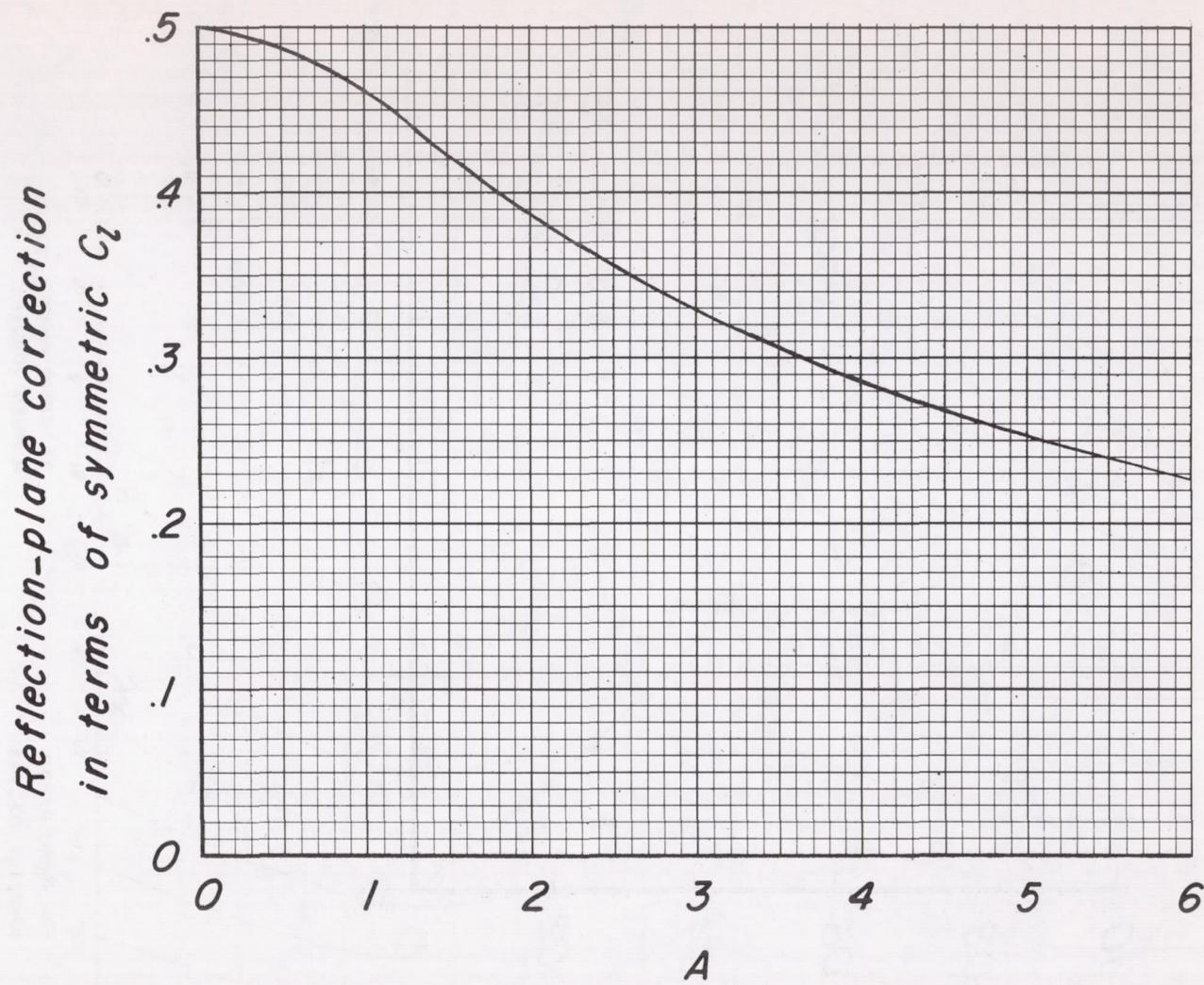


Figure 4.- Variation of reflection-plane correction with aspect ratio  
for full-span controls on untapered, unswept wings.

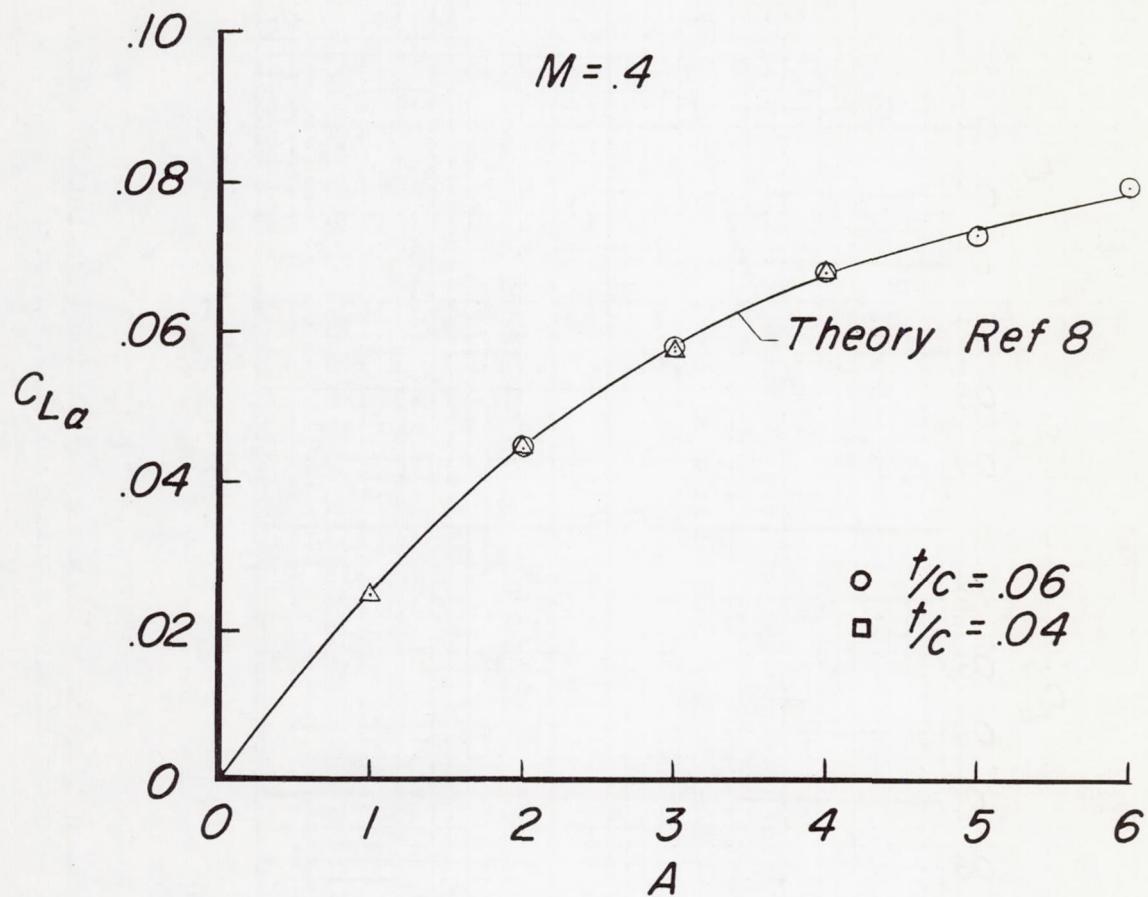


Figure 5.- Variation of the lift-curve slope with aspect ratio at a Mach number of 0.4.

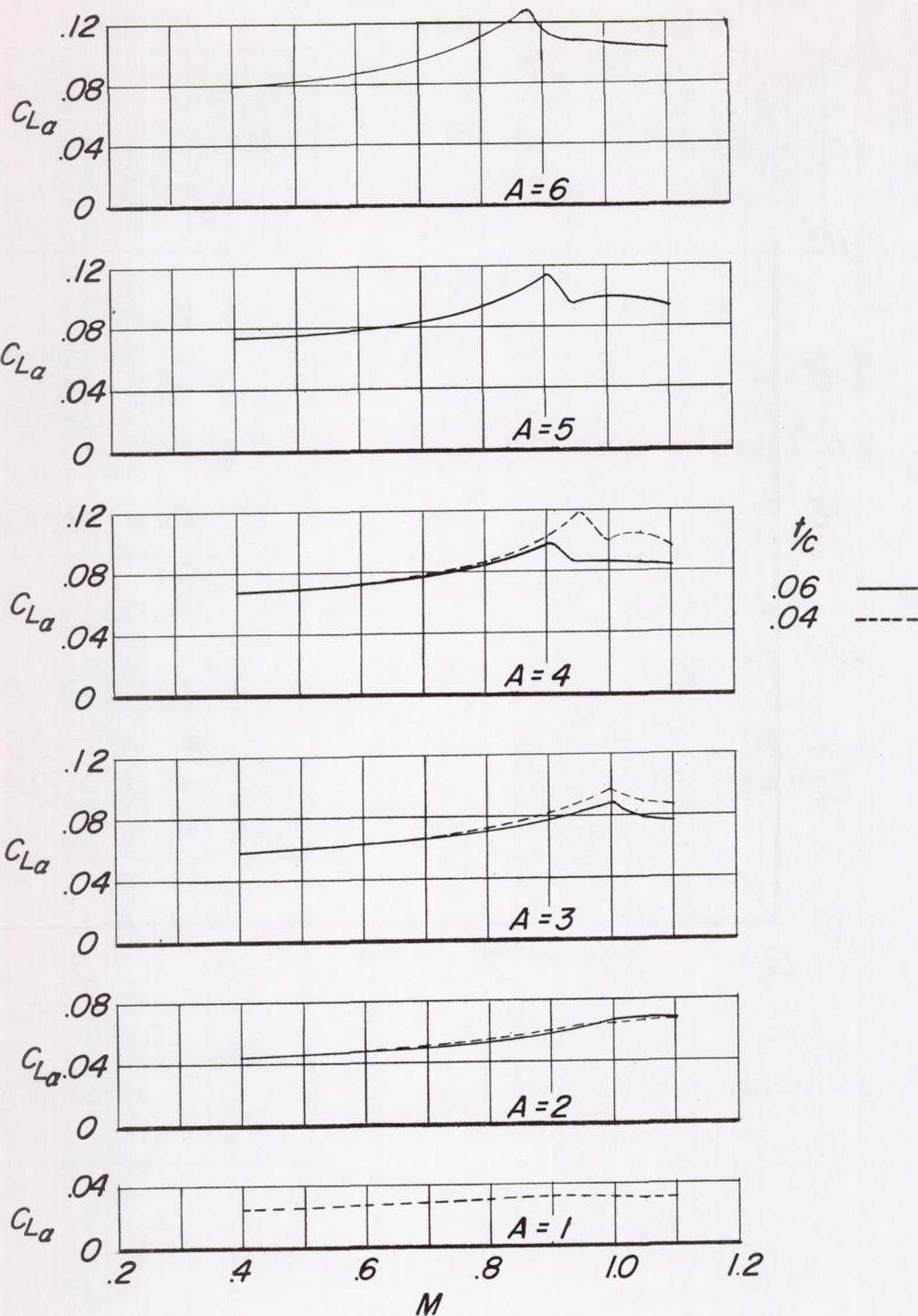


Figure 6.- The variation of lift-curve slope with Mach number.

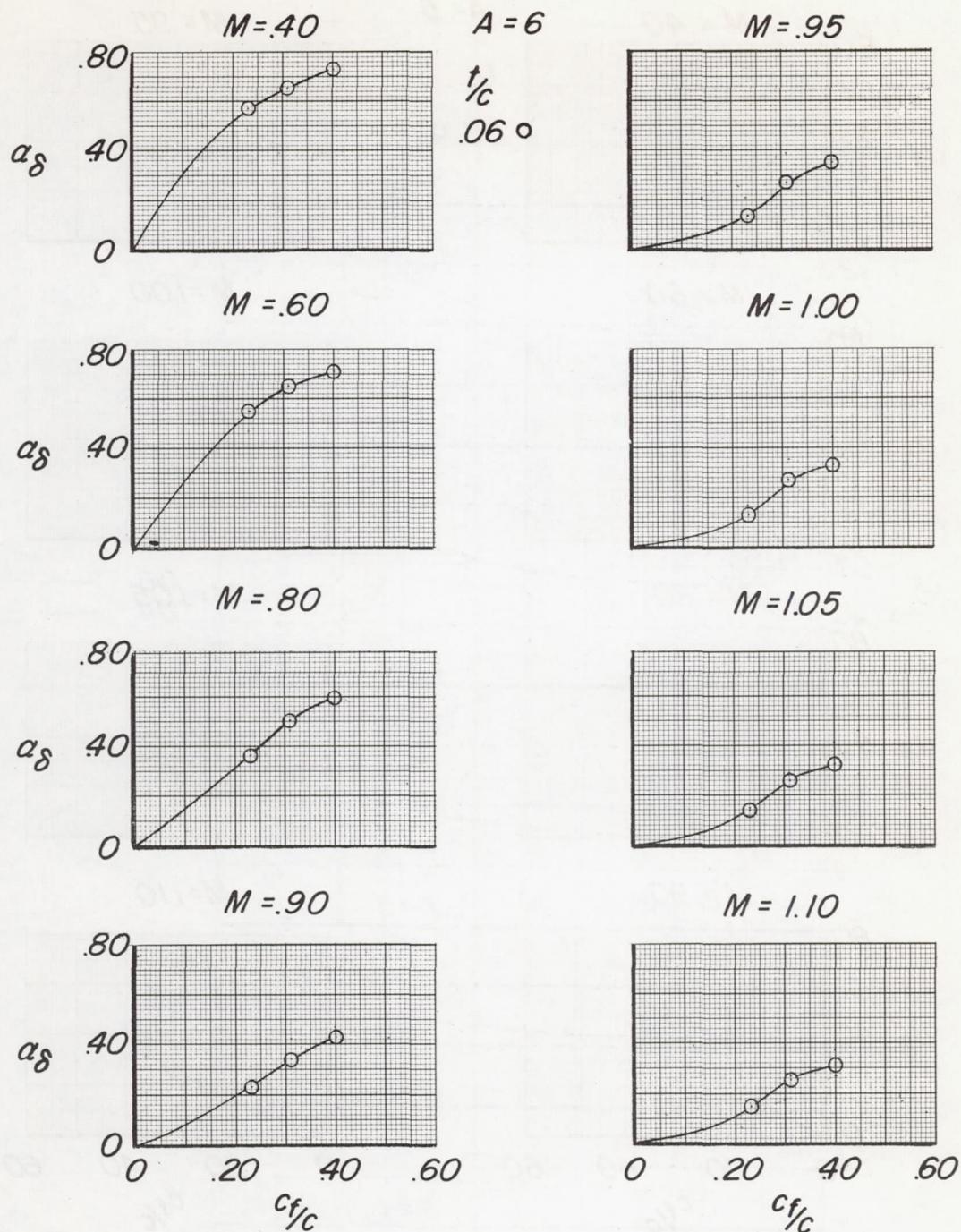
(a)  $A = 6$ .

Figure 7.- Flap-effectiveness parameter  $\alpha_d$  as a function of  $c_f/c$ .

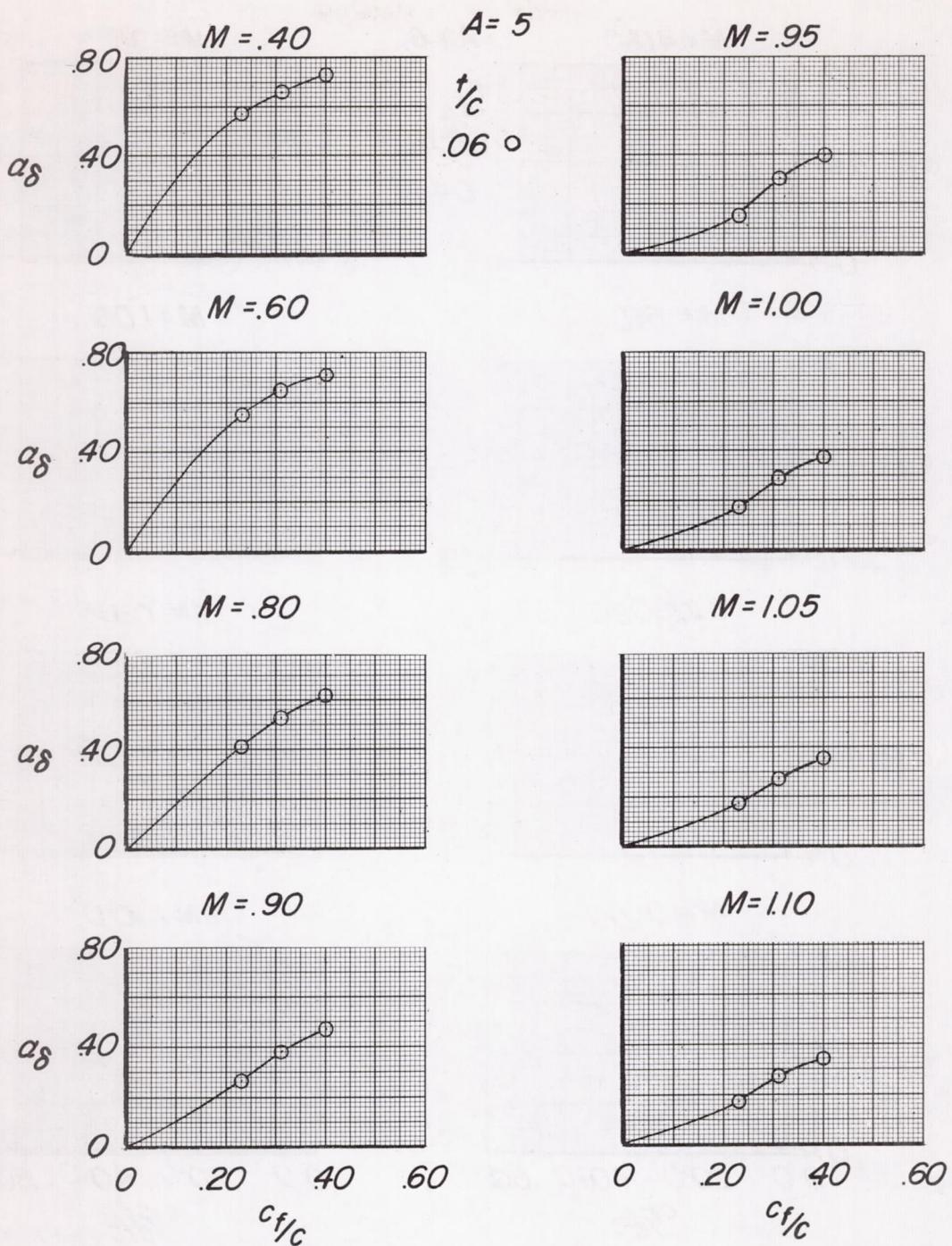
(b)  $A = 5.$ 

Figure 7.- Continued.

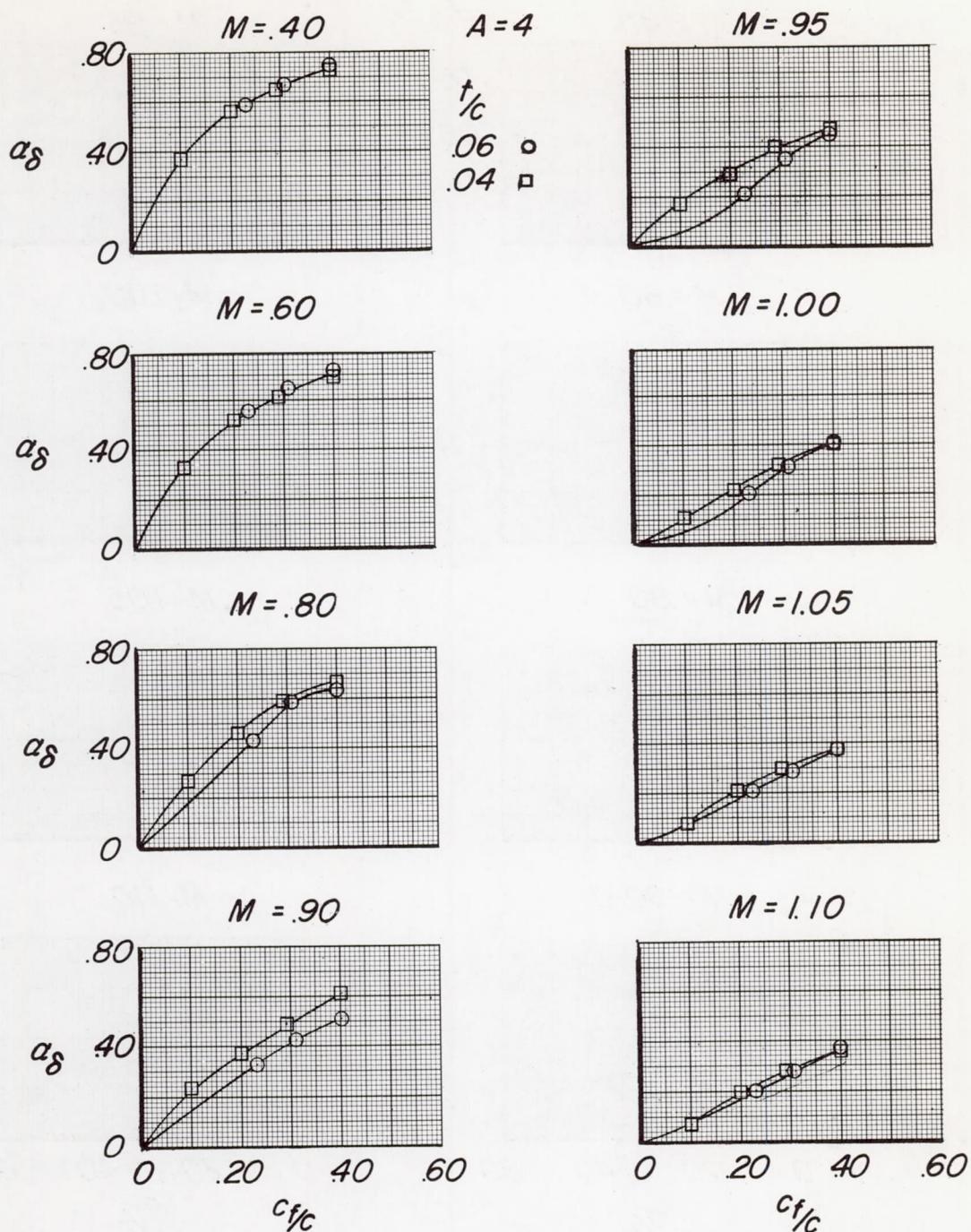
(c)  $A = 4.$ 

Figure 7.- Continued.

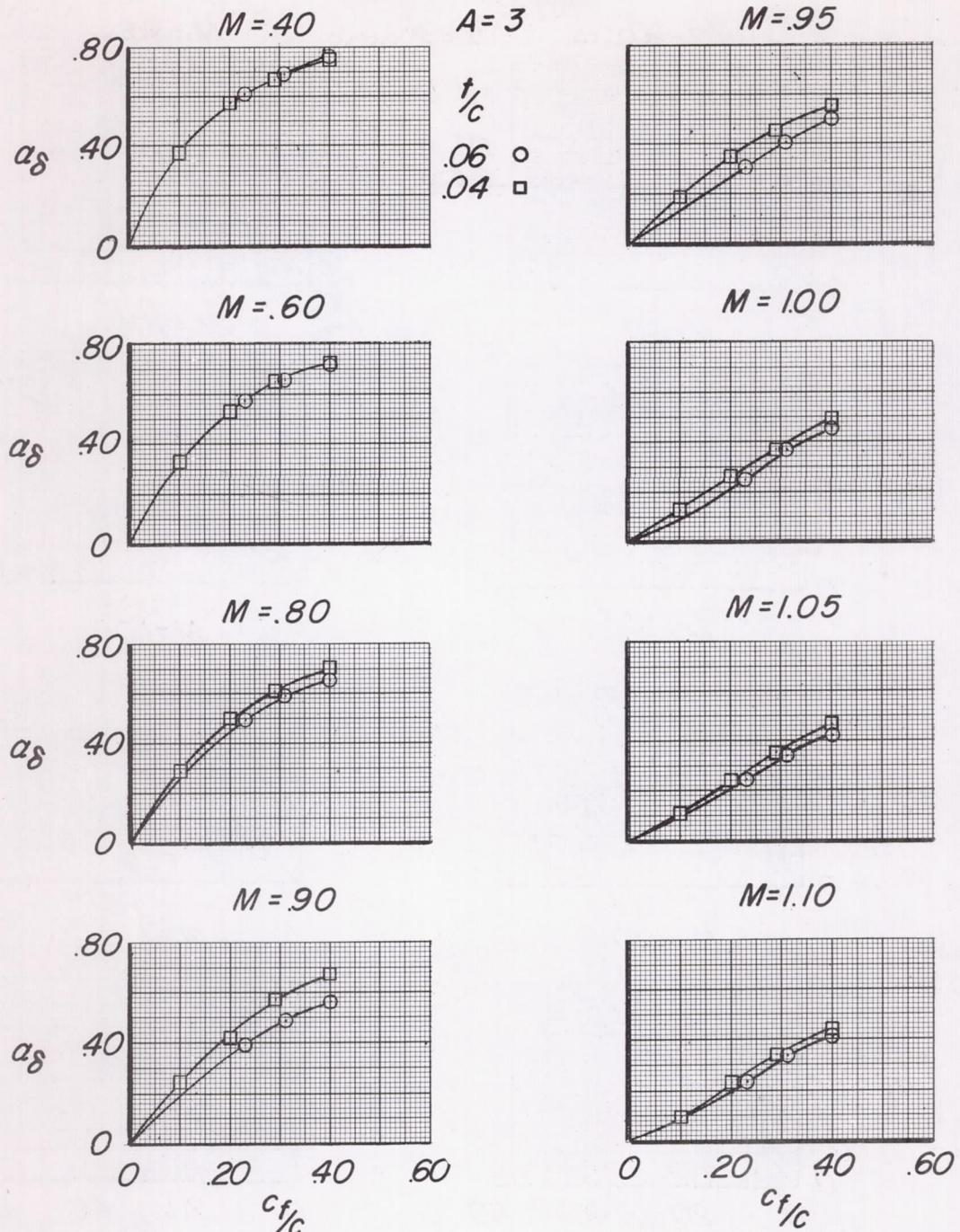
(d)  $A = 3.$ 

Figure 7.- Continued.

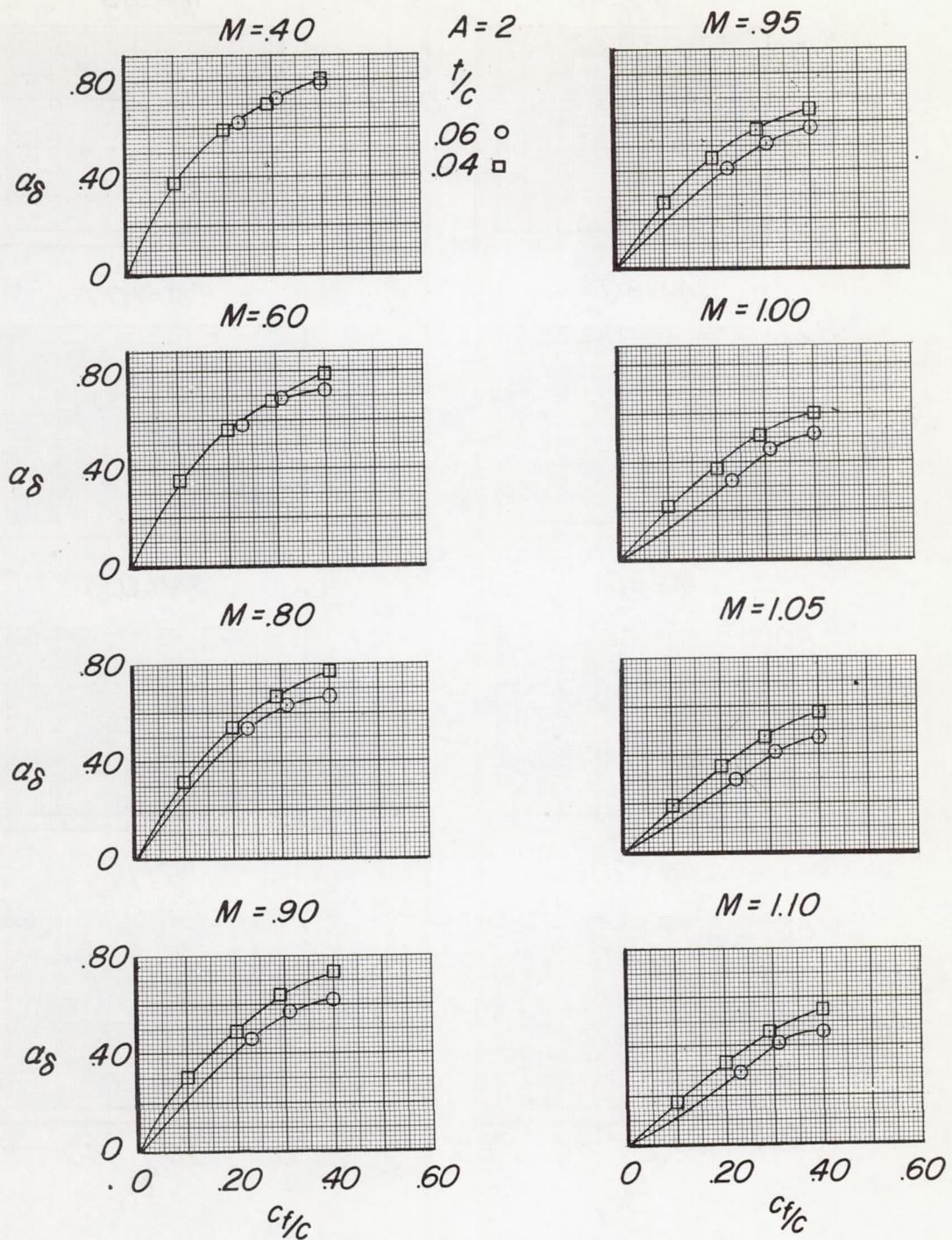
(e)  $A = 2.$ 

Figure 7.- Continued.

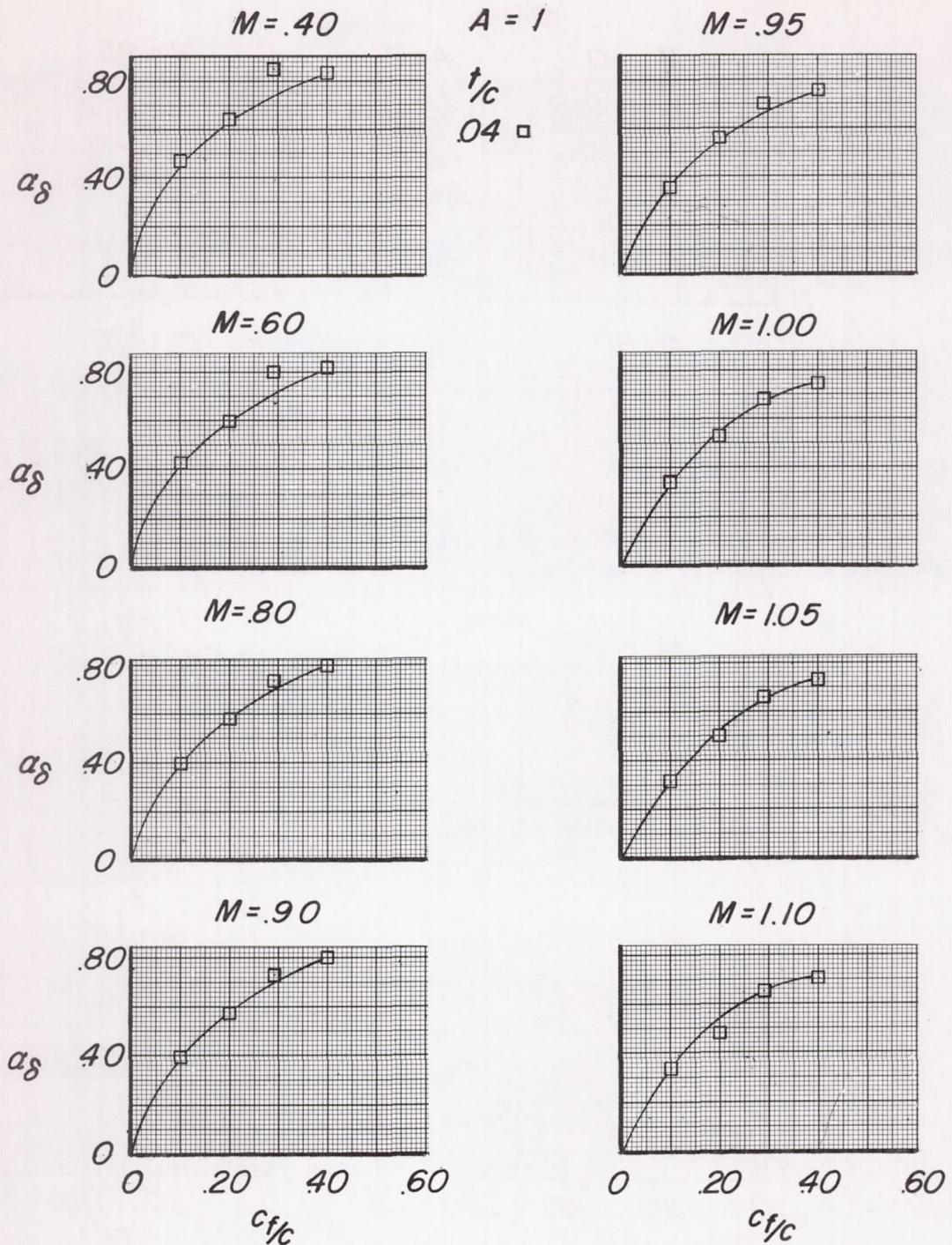
(f)  $A = 1.$ 

Figure 7.- Concluded.

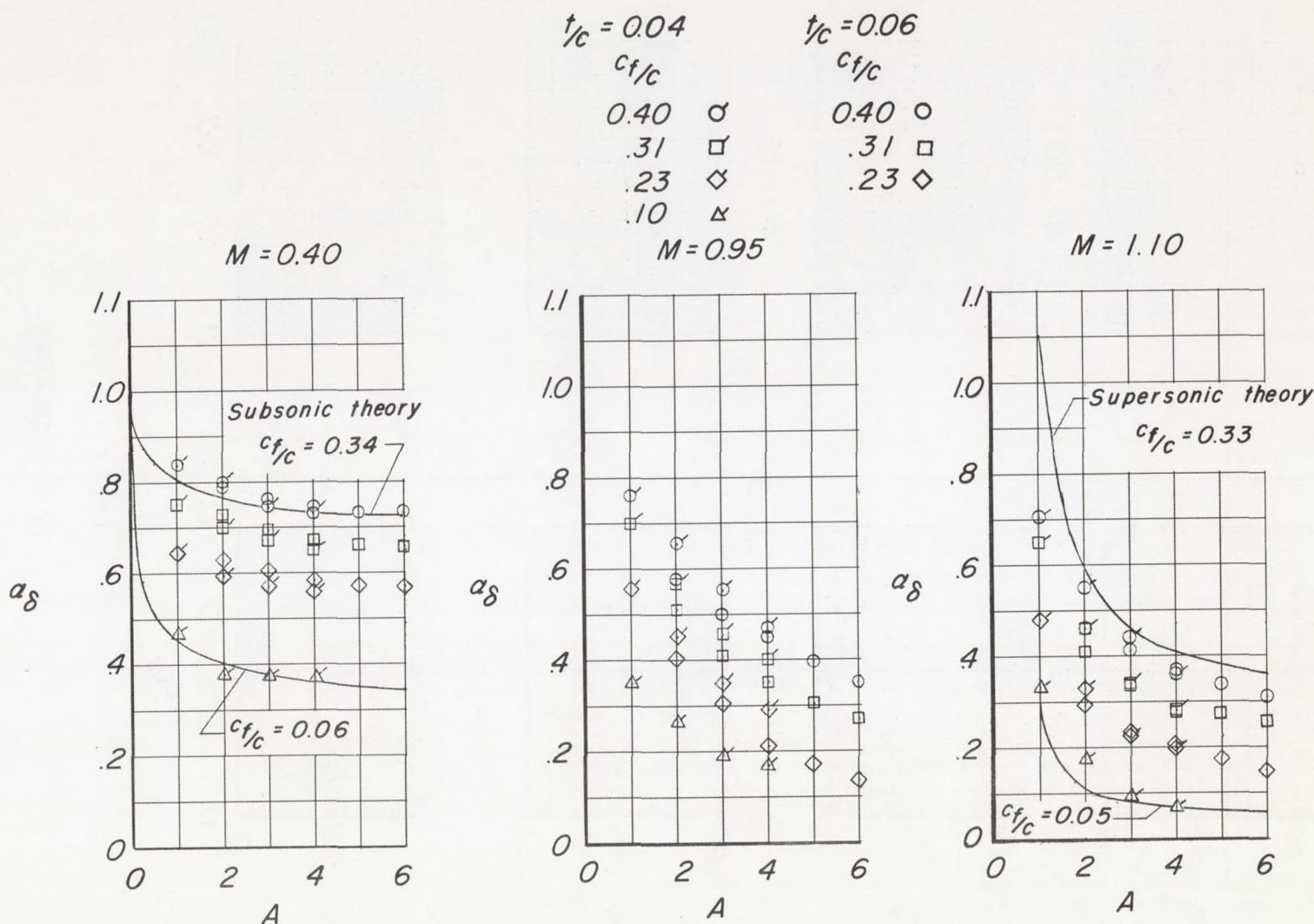


Figure 8.- Variation of  $\alpha_\delta$  with aspect ratio at various Mach numbers.