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SURFACE PRESSURE DISTRIBUTIONS ON A LARGE-SCALE 49° SWEPTBACK WING-BODY-TAIL CONFIGURATION WITH BLOWING APPLIED OVER THE FLAPS AND WING LEADING EDGE

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NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

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SUMMARY

An investigation has been conducted in the Langley full-scale tunnel to determine the pressure-distribution characteristics of a large-scale sweptback wing-body-tail configuration as affected by blowing air over highly deflected trailing-edge flaps in combination with several wing-leading-edge flow-control devices, including leading-edge blowing. The wing had an aspect ratio of 3.5, a taper ratio of 0.3, a leading-edge sweep of 49°, and NACA 65A006 airfoil sections parallel to the plane of symmetry. The tests were conducted over a range of angles of attack, flap angles, and aileron angles. The range of momentum coefficients for flap and leading-edge blowing varied from 0 to 0.187 and from 0 to 0.034, respectively. The Reynolds number of the tests was 5.2×10^6, corresponding to a Mach number of 0.08.

The results of this investigation showed that blowing at moderate momentum rates over highly deflected flaps increased the flap loads and induced some additional loading on the portion of the chord forward of the flap. Blowing over the flaps at high momentum rates resulted in very large flap loads and hinge moments and extremely high peak negative pressures over the flap leading edge. Blowing over a drooped leading edge from slots in the wing leading edge or knee resulted in very large values of wing-leading-edge normal-force and hinge-moment coefficient when used in combination with flap blowing. The type of wing-leading-edge flow-control device used and the rate of flap blowing had a large effect on the span loading but did not result in any unusual low-speed wing-root bending-moment problems.

INTRODUCTION

The need for increased lift capabilities of thin low-aspect-ratio wings, resulting from ever-increasing wing loadings, has led to considerable research on methods for attaining more lift on currently used wing geometries. Literature reporting the results of investigations along this line shows that boundary-layer control applied to a trailing-edge flap can produce appreciable incremental lift increases. However, if this greater flap effectiveness is to be exploited to the fullest extent on wings with thin airfoil sections, a more effective leading-edge-stall control system than is now used with conventional trailing-edge flaps must be employed.

A recent study of the leading-edge-stall control problem (ref. 1) demonstrated the need for using full-span devices on a thin swept wing equipped with trailing-edge-flap boundary-layer control, and limited tests utilizing blowing from a slot near the wing leading edge suggested the possibility of combining blowing with wing-leading-edge droop to provide effective stall control. The model of reference 1, therefore, was modified to incorporate a full-span leading-edge drooped nose of 0.17 chord provided with blowing slots to allow air to be ejected near the airfoil nose or at the knee of the droop. The results of the modified-wing tests (ref. 2) indicate that blowing from the knee of a drooped wing leading edge was an effective stall-control device.

In conjunction with the high-lift and stall-control investigations of references 1 and 2, complementary wing, flap, slat, droop, and aileron pressure distributions were recorded for some of the test configurations. This paper has been prepared to show the effects of wing-leading-edge-blowing and flap-blowing boundary-layer control on the chordwise pressure distributions and on the wing-leading-edge, flap, and aileron normal-force and hinge-moment characteristics.

The model used for these pressure-distribution tests was a large-scale 49° sweptback wing-body-tail configuration having an aspect ratio of 3.5, a taper ratio of 0.3, and NACA 65A006 airfoil sections parallel to the plane of symmetry.

Tests were conducted over a range of angles of attack, flap angles, and aileron angles. The range of momentum coefficients for flap and leading-edge blowing was 0 to 0.187 and 0 to 0.034, respectively. The Reynolds number of the tests was 5.2×10^6, corresponding to a Mach number of 0.08.

SYMBOLS

b	wing span, ft
c	local wing chord measured parallel to plane of symmetry, ft
$\bar{c}$	mean aerodynamic chord, $\frac{2}{S}\int_0^{b/2} c^2 dy$, ft
c_{av}	average chord of wing measured parallel to plane of symmetry, S/b, ft
p	local static pressure, lb/sq ft
p_∞	free-stream static pressure, lb/sq ft
q_∞	free-stream dynamic pressure, $\frac{1}{2}\rho_\infty {V_\infty}^2$
Q	volume flow of air ejected from slot, cu ft/sec
S	area of wing, sq ft
V_j	velocity of ejected air at slot, ft/sec
V_∞	free-stream velocity, ft/sec
x	chordwise distance measured parallel to plane of symmetry, ft
y	lateral distance from center line, measured perpendicular to plane of symmetry, ft
y_{cp}	distance from center line to lateral center of pressure, percent semispan
δ_a	aileron deflection (measured perpendicular to hinge line), deg
δ_f	flap deflection (measured perpendicular to hinge line), deg
ρ_j	mass density of air ejected from slot, slugs/cu ft
ρ_∞	mass density of free-stream air, slugs/cu ft

C_D drag coefficient (drag equivalent to pumping power not included), $\frac{\text{Drag}}{q_\infty S}$

C_L lift coefficient, $\frac{\text{Lift}}{q_\infty S}$

$C_{L,max}$ maximum lift coefficient

C_m pitching-moment coefficient about $\bar{c}/4$, $\frac{\text{Pitching moment}}{q_\infty S\bar{c}}$

C_μ momentum coefficient, $\frac{Q\rho_j V_j}{q_\infty S}$

C_l wing rolling-moment coefficient, $\int_0^1 \frac{c_n c}{c_{av}} \frac{2y}{b} d\,\frac{2y}{b}$

ΔC_l difference in rolling-moment coefficient produced by shutting off knee blowing on one wing (C_l with knee blowing on one wing - C_l with knee blowing off same wing)

C_N wing normal-force coefficient, $\int_0^1 \frac{c_n c}{c_{av}} d\,\frac{2y}{b}$

c_n wing section normal-force coefficient, $\int_0^1 c_p d\,\frac{x}{c}$

c_p pressure coefficient, $\frac{p - p_\infty}{q_\infty}$

$c_{n,a}$ aileron section normal-force coefficient based on and normal to aileron chord (up-load positive)

$c_{n,f}$ flap section normal-force coefficient based on and normal to flap chord (up-load positive)

$c_{n,LE}$ wing-leading-edge section normal-force coefficient based on and normal to leading-edge-droop chord (up-load positive)

$c_{n,s}$ slat section normal-force coefficient based on and normal to slat chord (up-load positive)

$c_{h,a}$ aileron section hinge-moment coefficient (moments about hinge line at 0.76c; negative value indicates a closing load for downward deflected aileron)

$c_{h,f}$ flap section hinge-moment coefficient (moments about hinge line at 0.76c; negative value indicates a closing load)

$c_{h,LE}$ wing-leading-edge section hinge-moment coefficient (moments about hinge line at 0.17c; negative value indicates a closing load)

$c_{h,s}$ slat section hinge-moment coefficient (moments about slat trailing edge; negative value indicates a closing load)

α angle of attack, deg

I inboard

C center

O outboard

Subscripts:

f flap

K knee

L left

LE leading edge

MODEL

A large-scale research model having the geometric characteristics shown in figure 1 was used in the investigation. The wing had a leading-edge sweep of 49°, an aspect ratio of 3.5, a taper ratio of 0.3, and NACA 65A006 airfoil sections parallel to the plane of symmetry. The model also had a horizontal tail mounted on the fuselage center line at a tail length of approximately $1.5\bar{c}$. A photograph of the model mounted for tests in the Langley full-scale tunnel is given as figure 2. Several wing leading-edge and trailing-edge flow-control devices were tested in this investigation. These include a full-span 0.17c leading-edge droop (fig. 3(a)), a 0.15c leading-edge slat extending from 0.40b/2 to the wing tip (fig. 3(b)), a 0.013c leading-edge modification that extended from the inboard end of the slat to the fuselage (fig. 3(c)), and 0.24c flaps (fig. 3(d)). The aileron geometry was the same as that of the flap but the blowing slot forward of the aileron was not used. Immediately forward of the flaps was an adjustable slot (see fig. 3(d)) through which air was ejected over the upper surface of the flap for boundary-layer control.

The leading-edge droop was equipped with blowing slots located either 1/2 inch from the wing leading edge or at the knee of the drooped leading edge. The knee slot became uncovered at a droop angle of 20° (normal to the droop hinge line). The leading-edge droop was divided into three spanwise sections - inboard, center, and outboard (which will be referred to as sections I, C, and O, respectively, in this report) for the purpose of regulating the spanwise extent and amount of leading-edge blowing.

APPARATUS

Chordwise pressure orifices were located on the upper and lower surfaces of the left wing at spanwise distances from the center line of 31, 56, 76, and 90 percent of the semispan - hereinafter referred to as stations 1, 2, 3, and 4, respectively. (See fig. 4.) Chordwise pressure orifices were also located on the upper surface of the leading-edge slat at spanwise distances of 56, 76, and 90 percent of the semispan (stations 2, 3, and 4, respectively). The surface static pressures were measured on a multiple-tube manometer and photographically recorded. The boundary-layer-control air-supply systems and flow-measurement procedures used in this investigation are fully described in references 1 and 2.

TESTS

Pressure-distribution data were obtained for a range of angles of attack to about 20°. The Reynolds number was 5.2×10^6, which corresponds to a Mach number of 0.08.

The configurations tested included (1) the basic wing with flaps undeflected and deflected, (2) the wing with slats or with slats and the inboard radius modification, with flaps deflected, and (3) the wing with leading edge drooped and flaps deflected. Trailing-edge boundary-layer-control tests were made for these configurations with a flap deflection angle of 60°.

The leading-edge-blowing test conditions included various combinations of blowing from compartments I, C, and O at the nose or knee slot locations. For the drooped-leading-edge configuration with flap blowing applied, pressure-distribution data were also obtained for a range of aileron deflections from 0° to 40° without and with knee blowing applied.

DATA REDUCTION

The measured surface static pressures were reduced to coefficient form and plotted against their respective chordwise locations. The section normal-force and hinge-moment coefficients obtained from the pressure-distribution data were determined by mechanical integration processes. With flap blowing applied, the flap chord forces were included in the appropriate calculations. For the tests without flap blowing, however, the flap chord forces were determined to be negligible and were not included in the calculations. The pressure-distribution data were corrected for the average effects of airstream misalinement and jet-boundary effects on the angle of attack.

RESULTS AND DISCUSSION

Presentation

The results of the investigation have been presented in both tabular and graphical form. The measured surface pressure-coefficient data are contained in tables 1 to 33. Some of the more pertinent information describing the first-order effects of blowing boundary-layer control are presented in graphical form as chordwise pressure distributions, spanwise loadings, normal forces, and hinge moments. In the following table the figures are listed according to type of data and wing-leading-edge configuration:

	Figures
Chordwise pressure distributions:	
Basic	5 to 8
Slatted	9 to 11
Drooped with aileron neutral	12 to 16
Drooped with aileron deflected	17 to 19

Wing Chordwise Pressure Distributions

Basic wing leading edge; effect of flap deflection and flap blowing without and with blowing over wing leading edge.- The wing chordwise pressure distributions for the basic-leading-edge configuration (clean, undeflected leading edge with original airfoil contour) without flap blowing applied over undeflected and deflected flaps are shown for a range of angles of attack in figure 5. The flap is stalled for both the 40° and 60° flap deflections throughout the angle-of-attack range; therefore the pressure diagrams for these two flap deflections are similar. The deflected flap (stations 1 and 2) does, however, carry a larger load than the nondeflected flap, and induces some additional loading just forward of the flap.

The effect on the pressure distributions of applying flap blowing over 60° deflected flaps and the effect of applying wing-leading-edge blowing in combination with the flap blowing are shown in figure 6. Blowing over the flap at the low momentum coefficient $C_{\mu,f}$ of 0.014 increased the loading over the flap through the angle-of-attack range. (Compare figs. 5(c) and 6(a).) Increasing the flap blowing rate resulted in additional flap loading. (Compare figs. 6(a) and 6(c).)

The data of figures 5 and 6 and references 1 and 2 show that maximum lift for the basic wing-leading-edge configuration with flap blowing applied is limited because of wing-leading-edge separation at relatively low angles of attack. In an attempt to delay this leading-edge separation and thereby improve the chordwise loadings, full-span wing-leading-edge blowing was applied in combination with a high flap momentum coefficient $C_{\mu,f}$ of 0.102. The pressure distributions (fig. 6(c)) show a significant increase in loading over the outboard wing sections for the angle-of-attack range presented. The most important chordwise-loading aspect of leading-edge blowing, the high angle-of-attack range, unfortunately was not obtained; however, the data of reference 2 show that leading-edge blowing over the basic wing leading edge was not sufficient to control wing-leading-edge separation beyond about 9° angle of attack. (See fig. 8.)

The general effects of flap blowing and wing-leading-edge blowing on the chordwise pressure distributions just discussed are illustrated more clearly in figure 7, which shows the chordwise pressure distributions at stations 2 and 4 without and with flap blowing and leading-edge blowing applied for an angle of attack of approximately 9°. Flap blowing resulted in a large increase in loading over the flap (fig. 7(a)) and also resulted in an increase in loading over the outboard portion of the wing as well as the aileron (fig. 7(b)). Leading-edge blowing in conjunction with flap blowing resulted in additional loading over the whole wing chord for the angle of attack presented. For convenience, the force and moment data as determined from references 1 and 2 are presented in figure 8 for the wing configurations just discussed.

Slatted wing leading edge; effect of flap blowing and wing-leading-edge radius increase.- The chordwise pressure distributions of the wing with 0.60b/2 slats installed, without and with flap blowing applied, are shown in figure 9. Also shown in figure 9 are the chordwise pressure distributions resulting from modification of the wing leading edge by a radius increase (fig. 3(c)) from the inboard end of the 0.60b/2 slat to the fuselage.

The effects of flap blowing and the radius increase for an angle of attack of about 13° are shown for stations 1 and 2 in figure 10. The force and moment data from references 1 and 2 corresponding to the pressure-distribution data just discussed are presented in figure 11.

The slat without flap blowing applied (fig. 9(a)) controlled wing-leading-edge separation through most of the angle-of-attack range presented. The only regions of leading-edge-separated flow indicated are inboard of the slat (station 1) at angles of attack of 13.5° and 17.3° and at the tip (station 4) for an angle of attack of 17.3°. These regions of separation were not very detrimental to lift, however, because the deflected-flap lift increment, shown in figure 11, remained essentially constant through the complete angle-of-attack range. With flap blowing applied, however, the wing-leading-edge separated flow (inboard of the slat) was determined from flow studies to be very severe and to pass rearward and directly over the flap. (See ref. 1.) The flow separation resulted in reduced flap effectiveness with increasing angle of attack (figs. 9(b) and 11).

Previous experience has shown that the inboard leading-edge flow separation could be relieved or eliminated by increasing the leading-edge radius inboard of the slat. (See fig. 3(c).) The chordwise pressure distributions resulting from this modification are shown in figures 9(c) and 10(a). The inboard separation was nearly eliminated (fig. 9(c) and ref. 1) and the flap loads were increased (fig. 10(b)). Although the pressure data of figure 9(c) did not include the high angle-of-attack range, the flow studies and force data of reference 1 show that the maximum lift characteristics were improved by the alleviation of the inboard wing-leading-edge separation.

Drooped wing leading edge; effect of flap blowing without and with blowing over wing leading edge or knee.- The chordwise pressure distributions of the wing with the leading edge drooped 45° and with blowing applied over the leading edge or knee and/or flap are shown in figure 12. The pressure distributions at stations 2 and 4 for an angle of attack of about 17° are shown in figure 13. The drooped leading edge alone was not sufficient to control separation at the knee of the droop for the two highest angles of attack shown (fig. 12(a)). Wing-leading-edge or knee blowing were, however, about equally effective in controlling separation to high angles of attack (figs. 12(b) and 12(c)). The main effect of leading-edge or knee blowing (fig. 13) was to increase the chordwise loadings at high angles of attack, and it was evident from the large increases in wing-leading-edge loading that such flow-control devices were about equally effective in eliminating the leading-edge-stall problem for the angle-of-attack range investigated.

First-order effects on the chordwise pressure distributions of blowing air over trailing-edge flaps at relatively high momentum rates ($C_{\mu,f} \approx 0.18$) were obtained for the 40° drooped leading-edge configuration. This momentum rate is equivalent to a weight flow rate of the order of 40 to 50 pounds of air per second and represents a blowing condition which utilizes, for lift augmentation, about 40 percent of the total engine thrust of a present-day J-79 engine operating at 85 percent revolution speed. The pressure-distribution results of these tests are shown in figure 14. In general, wing-leading-edge or knee blowing in combination with the droop controlled the wing-leading-edge separation through the angle-of-attack range investigated. It should be noted that very large flap loads resulted from the high flap-blowing rate. Chordwise pressure distributions of figure 14(c) for station 2 at $\alpha = 15.9^\circ$, replotted in figure 15, show that section flap loads about 1/3 to 1/2 the total section loading resulted from the high flap-blowing rate. Furthermore, the peak negative pressures over the flap leading edge were very large (c_p values of about -26).

The force and moment test results of references 1 and 2 corresponding to the pressure-distribution data of figures 12 to 15 are presented in figure 16.

Drooped wing leading edge; effect of aileron deflection and flap blowing without and with knee blowing.- Wing chordwise pressure distributions are presented in figure 17 for several aileron deflection angles over a range of angles of attack, without and with knee blowing applied in combination with flap-blowing momentum coefficient of approximately 0.10. The aileron chord loading (stations 3 and 4) at corresponding angles of attack increased with increasing aileron deflection angle through about 30° with knee blowing applied, but for 40° deflection a decrease in loading is noted. Despite this decrease in loading, the limited leading-edge-blowing data of reference 1 showed that the aileron effectiveness remained good through 40° deflection. Therefore, the aileron loading is not wholly responsible for the indicated aileron effectiveness. The chordwise pressure distributions of figure 17, together with the large-scale plots of figures 18 and 19, show that knee blowing resulted in a large increase in load forward of the aileron for all deflection angles.

The data of figures 17, 18, and 19 indicate that rolling effectiveness could probably be obtained, even with drooped ailerons, by varying the rate or spanwise extent of wing-leading-edge blowing (knee blowing in this case) and thereby controlling the loading of the whole outboard portion of the wing. A more detailed analysis of the use of leading-edge blowing for lateral control is given in the following section.

Wing Spanwise Loadings

Wing-leading-edge configurations with aileron neutral.- The span-loading characteristics of several of the previously discussed wing configurations are presented in figure 20 for a range of angles of attack. In that the most inboard pressure-distribution station was 0.31b/2, several possibilities exist for extending the span loading curves to the center line. The curves used inboard of the 0.31b/2 station resulted from consideration of the probable fuselage loads and the type of loading believed to exist across the fuselage-wing juncture and the flap-end discontinuity. The resulting normal-force coefficients were within 10 percent of the wing normal-force coefficients as determined from the force data. The data of figure 20 show that the type of wing-leading-edge flow-control device and the rate of flap blowing have a large effect on the wing span-loading characteristics.

The lateral variations of the centers of pressure of the wing configurations shown in figure 20 are presented as a function of lift coefficient in figure 21. The lateral center of pressure was at about 0.42b/2 over the C_L range and the maximum lateral movement of the center of pressure was about 5 percent through the C_L range for the high-lift configurations. In no case did the center-of-pressure movement exceed that of the basic-leading-edge configuration; therefore, wing-leading-edge, knee, or flap blowing should not cause any unusual wing-root bending-moment problems at low speeds.

Drooped wing leading edge with aileron deflected.- The span-loading characteristics of the wing configuration with 40° drooped leading edge, without and with knee blowing applied in combination with flap blowing, are shown in figure 22 for several aileron deflection angles. Knee blowing is seen to improve the loading over the outboard portion of the wing for the higher angles of attack. This increase in loading over the outboard wing sections did not cause any significant change in the spanwise center of pressure as seen in figure 23; however, because of the larger values of normal-force coefficient associated with the knee blowing, the wing rolling-moment characteristics were appreciably affected. (See fig. 24.)

The incremental values of C_l shown in figure 24 are the difference between the rolling-moment coefficients without knee blowing and with knee blowing applied, obtained by integrating the span-loading curves of figure 22 about the lateral plane of symmetry. The data of figure 24, therefore, are equivalent to a blowing system in which the knee blowing was eliminated for one wing semispan. The significant portion of the data of figure 24 is that for angles of attack of approximately 12° or greater because the momentum-coefficient values presented would be applicable to an airplane only at low speed (i.e., at relatively high angles of attack). For angles of attack of 12° or greater very large values of rolling-moment coefficient resulted from shutting off the air from the knee on one wing. For an airplane with the same geometric characteristics as the present model, values of ΔC_l of 0.12 would result in a rolling effectiveness parameter $pb/2V_\infty$ about 3.5 times the normal value of 0.09. Sufficient rolling effectiveness could probably be obtained by only partially throttling the air from the knee on one wing. The control of the amount of air and the resultant response rates for roll control, however, would have to be established by more extensive tests.

Normal Forces and Hinge Moments

Probably the most significant aspect of the present paper from the viewpoint of loads and design is included in the following section, which is concerned with loads and hinge moments of component parts of the wing as affected by wing-leading-edge, knee, or flap blowing. This portion of the paper provides substantial design information appropriate for the current speed range of take-off and landing, wherein the controls and wing-leading-edge devices would be at large deflection angles.

Basic wing leading edge; effect of flap deflection and flap blowing without and with blowing over wing leading edge.- The wing-leading-edge, flap, and aileron section normal-force and hinge-moment coefficients for the basic wing-leading-edge configuration are shown in figures 25 and 26 for a range of flap deflections and for several flap-blowing rates. For the high flap-blowing rate $C_{\mu,f}$ of 0.102, full-span wing-leading-edge blowing was also applied. For the inboard stations (1 and 2), figure 25(a) shows that the values of $c_{n,LE}$ are, in general, increased in the low angle-of-attack range with either increasing flap deflection angle or flap-blowing rate. Without or with flap blowing applied, separation occurred at relatively low angles of attack and resulted in a rapid decrease in the values of $c_{n,LE}$. Wing-leading-edge blowing provided larger values of $c_{n,LE}$ at the two outboard stations than flap blowing alone; however, flow separation and the resulting loss in $c_{n,LE}$ occurred at about the same angle of attack as without leading-edge blowing.

The wing-leading-edge hinge-moment coefficients (fig. 26(a)) became large at one or more stations in the low to moderate angle-of-attack range ($\alpha = 6^{\circ}$ to 12°) for all the test conditions. Leading-edge blowing increased the values of $c_{h,LE}$ at station 4 to about four times those experienced with flap blowing alone at an angle of attack of about 9°.

The flap and aileron section normal-force and hinge-moment coefficients (figs. 25(b) and 26(b)) were, in general, increased with increasing flap-blowing rate for a given wing station and angle of attack. The only significant effect of wing-leading-edge blowing on the flap and aileron normal-force or hinge-moment coefficients was to increase the aileron normal-force coefficients through the angle-of-attack range.

<u>Drooped wing leading edge with aileron neutral; effect of flap blowing without and with blowing over wing leading edge or knee</u>.- The wing-leading-edge, flap, and aileron section normal-force and hinge-moment coefficients for the wing configuration with drooped leading edge are shown in figures 27 and 28, without and with flap blowing in combination with wing-leading-edge or knee blowing. Wing-leading-edge or knee blowing is seen to result in large values of wing-leading-edge normal-force and hinge-moment coefficients through the angle-of-attack range investigated.

The flap normal-force coefficients obtained with wing leading-edge or knee blowing were four to five times as large as those experienced with only flap blowing applied (fig. 27(b)); however, the hinge-moment coefficients were not significantly affected (fig. 28(b)). The aileron normal-force coefficients obtained with leading-edge or knee blowing applied were about three times as large as those resulting from flap blowing alone, but the associated hinge-moment coefficients were not significantly affected by wing-leading-edge, knee, or flap blowing.

The effects on the wing-leading-edge, flap, and aileron section normal-force and hinge-moment coefficients caused by blowing over the flaps alone at a very high momentum rate ($C_{\mu,f} \approx 0.18$) and in combination with wing-leading-edge or knee blowing are shown in figures 29 and 30. The values of $c_{n,LE}$ and $c_{h,LE}$ (figs. 29(a) and 30(a)) become very large with flap blowing applied (maximum values of about 3.6 and -2.2, respectively) and wing-leading-edge or knee blowing further increases these coefficients to maximum values of about 6.0 and -3.4, respectively. Such large values of wing-leading-edge normal-force and hinge-moment coefficients would be expected to require special consideration in the design of the leading-edge droop and supporting brackets and structures. The flap normal-force and hinge-moment coefficients (figs. 29(b) and 30(b)) are also large as a result of the high flap-blowing rate and will also require special consideration by the loads designer.

Drooped wing leading edge with aileron deflected; effect of flap blowing without and with knee blowing.- Aileron section normal-force and hinge-moment coefficients are presented for several aileron droop angles in figures 31 and 32, respectively, for a range of angles of attack, without and with knee blowing applied in combination with a flap-blowing momentum coefficient of about 0.10. For moderate angles of attack the aileron normal-force coefficients (fig. 31) increased as the deflection angle was increased to 30°, without or with knee blowing. Knee blowing resulted in a large increase in aileron normal-force coefficient for a deflection angle of 30°. Increasing the angle of attack above about 10°, however, resulted in a considerable decrease in aileron loading.

The aileron section hinge-moment coefficients (fig. 32) without knee blowing applied were increased fairly uniformly with increasing aileron deflection angle, without much effect related to angle of attack, in the low to moderate angle-of-attack range. Above an angle of attack of about 10° the hinge-moment coefficients varied considerably with angle of attack and deflection angle because of section stalling. With knee blowing applied the hinge-moment coefficients were, in general, increased uniformly with increasing deflection angle, and increasing angle of attack had only a minor effect on these hinge moments.

CONCLUSIONS

Results of tests of blowing boundary-layer control in the Langley full-scale tunnel to determine the loads on the wing and high-lift devices of a large-scale 49° sweptback wing-body-tail configuration indicate the following conclusions:

1. Wing-leading-edge or knee blowing, when used in combination with flap blowing at high momentum rates, resulted in wing-leading-edge normal-force and hinge-moment coefficients so large that special consideration of these values will be required by the loads designer.

2. Flap blowing at low momentum-coefficient rates (<0.02) increased the flap loads and induced some additional loading forward of the flaps. The flap loads at these low momentum rates, however, were seriously reduced when separated flow existed forward of the flap.

3. Flap blowing at the high momentum-coefficient rate of 0.18 resulted in flap loads that were from one-third to one-half of the wing-flap section load and the peak negative pressure coefficients over the flap leading edge were very large (about -26), and thus also require special consideration by the loads designer.

4. The type of wing-leading-edge flow-control device used and the rate of flap blowing over the flaps had a large effect on the span-loading characteristics of the wing.

5. The lateral center-of-pressure movement with leading-edge and/or flap blowing was in general less than 5 percent through the test angle-of-attack range, and in no case was greater than that for the basic unflapped configuration; therefore, wing-leading-edge or flap blowing should not cause any unusual wing root bending-moment problems at low speed.

6. Throttling the knee blowing on one wing semispan resulted in very large rolling-moment coefficients for angles of attack greater than 12°.

Langley Aeronautical Laboratory,
National Advisory Committee for Aeronautics,
Langley Field, Va., Nov. 4, 1957.

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TABLE 1

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 00^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu f} =$ .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 6.8°				α = 10.3°			
WING									
Upper	.0000	-3.171	-4.498	-1.517	-1.332	-6.748	-1.665	-.916	-.621
	.0025	-2.705	-4.189	-1.440	-1.164	-4.367	-1.969	-.890	-.567
	.0050	-2.434	-3.174	-1.297	-1.176	-3.767	-1.565	-.840	-.567
	.0075	-2.213	-2.672	-1.282	-1.191	-3.252	-1.565	-.844	-.567
	.0100	-2.016	-2.363	-1.290	-1.187	-2.840	-1.569	-.848	-.567
	.0150	-1.581	-1.985	-1.344	-1.168	-2.367	-1.577	-.852	-.563
	.0250	-1.124	-1.483	-1.320	-1.172	-2.046	-1.611	-.852	-.563
	.0400	-.833	-1.139	-1.332	-1.145	-1.947	-1.642	-.855	-.529
	.0800	-.624	-.776	-1.290	-1.160	-1.241	-1.735	-.840	-.529
	.1300	-.473	-.583	-1.247	-1.168	-.653	-1.554	-.802	-.521
	.1700	-.442	-.544	-1.139	-1.122	-.603	-1.308	-.802	-.506
	.2800	-.357	-.425	-.583	-.660	-.489	-.800	-.760	-.467
	.4300	-.250	.089	-.297	-.225	-.256	-.370	-.734	-.429
	.5800	-.240	-.236	-.166	-.111	-.190	-.250	-.586	-.383
	.7300	-.100	-.050	-.031	.004	-.120	-.131	-.300	-.337
Lower	.0025	-.233	.010	.062	-.324	-1.641	-.858	-.049	-.218
	.0050	.186	.023	.239	.218	-.534	-.431	.133	.203
	.0100	.399	.301	.378	.408	.107	.196	.323	.360
	.0250	.430	.440	.432	.443	.454	.427	.449	.448
	.0400	.395	.344	.405	.386	.481	.404	.449	.410
	.0700	.329	.282	.340	.347	.447	.354	.418	.421
	.1700	.186	.135	.189	.183	.309	.242	.278	.226
	.2800	.151	.066	.116	.107	.240	.165	.182	.126
	.4300	.090	.031	.050	.019	.130	.127	.106	.050
	.5800	.089	.000	-.004	-.031	.115	.058	.027	-.031
	.7200	.016	.000	-.023	-.073	.057	.062	-.015	-.111
FLAP or AILERON									
Upper	.7310	.008	.066	.015	.008	.050	.100	.023	-.111
	.7395	.023	.069	.027	.011	.076	.100	.023	-.107
	.7532	-.070	-.050	.050	.073	-.099	-.108	-.319	-.299
	.7806	-.070	-.069	-.039	-.008	-.080	-.108	-.304	-.287
	.8354	-.021	-.023	-.004	.023	-.027	-.062	-.255	-.268
	.9038	.035	.027	.031	.027	.038	.000	-.205	-.245
	.9440	.074	.062	.035	.027	.076	.031	-.163	-.245
	.9863	.171	.097	.058	.027	.080	.069	-.129	-.241
	.9928	.180	.093	.058	.027	.111	.069	-.133	-.241
Lower	.7312	.000	.000	-.066	-.103	.061	.000	-.030	
	.7367	.023	-.050	-.050	-.088	.061	.038	-.027	-.119
	.7428	.039	-.050	-.031	-.084	.084	.058	-.011	-.100
	.7532	.039	-.050	-.035	-.080	.088	.077	-.011	-.103
	.7806	.035	-.025	-.023	-.031	.088	.058	-.004	-.080
	.8628	.070	.042	.023	-.008	.092	.058	-.004	-.080
	.9313	.089	.069	.062	.034	.115	.085	-.008	-.080

TABLE 1 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f =$ 00° $\delta_a =$ 00° $i_t =$ 00°

$C_{\mu f} =$ .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 14.0°				α = 17.8°			
WING									
Upper	.0000	-2.411	-1.178	-.611	-.448	-2.277	-.960	-.611	-.472
	.0025	-2.352	-1.591	-.631	-.424	-2.145	-1.191	-.611	-.448
	.0050	-2.305	-1.130	-.599	-.424	-2.032	-.936	-.583	-.456
	.0075	-2.313	-1.161	-.599	-.424	-2.032	-.940	-.587	-.456
	.0100	-2.324	-1.152	-.599	-.424	-2.008	-.944	-.587	-.456
	.0150	-2.320	-1.152	-.599	-.424	-2.048	-.944	-.587	-.456
	.0250	-2.348	-1.174	-.595	-.432	-2.108	-.944	-.587	-.452
	.0400	-2.384	-1.161	-.595	-.392	-2.189	-.944	-.587	-.420
	.0800	-2.712	-1.204	-.595	-.396	-2.024	-.948	-.575	-.424
	.1300	-2.300	-1.152	-.563	-.396	-1.800	-.908	-.556	-.424
	.1700	-1.980	-1.152	-.563	-.396	-1.875	-.908	-.556	-.424
	.2800	-.684	-1.126	-.524	-.396	-1.651	-.884	-.552	-.420
	.4300	-.420	-1.050	-.508	-.376	-.851	-.850	-.520	-.404
	.5800	-.304	-.948	-.508	-.376	-.618	-.821	-.496	-.404
	.7300	-.150	-.735	-.508	-.364	-.350	-.765	-.496	-.396
Lower	.0025	-1.368	-.852	-.095	-.236	-1.558	-.857	-.242	-.388
	.0050	-.561	-.513	.063	.128	-.771	-.598	-.067	-.016
	.0100	.043	.109	.274	.300	-.092	.024	.179	.212
	.0250	.439	.440	.433	.420	.410	.382	.385	.372
	.0400	.510	.491	.440	.412	.502	.430	.436	.404
	.0700	.506	.457	.421	.412	.554	.442	.448	.444
	.1700	.407	.339	.310	.232	.482	.371	.349	.304
	.2800	.324	.261	.222	.152	.410	.295	.266	.208
	.4300	.119	.191	.119	.052	.185	.227	.163	.100
	.5800	.150	.091	.004	-.040	.165	.100	.036	.000
	.7200	.071	.057	-.091	-.140	.072	.024	-.071	-.108
FLAP or AILERON									
Upper	.7310	.079	.100	-.040	-.140	.060	.072	-.032	-.128
	.7395	.099	.096	-.020	-.132	.060	.068	-.012	-.124
	.7532	-.107	-.730	-.532	-.408	-.398	-.773	-.544	-.436
	.7806	-.126	-.650	-.500	-.380	-.365	-.661	-.512	-.400
	.8354	-.067	-.544	-.488	-.340	-.297	-.653	-.500	-.380
	.9038	.004	-.417	-.464	-.328	-.185	-.574	-.476	-.372
	.9440	.020	-.317	-.450	-.312	-.145	-.562	-.452	-.376
	.9863	.070	-.269	-.440	-.308	-.100	-.530	-.460	-.348
	.9928	.103	-.239	-.425	-.308	-.052	-.502	-.460	-.344
Lower	.7312	.075	.044	-.103	-.168	.060	.004	-.091	-.148
	.7367	.075	.044	-.099	-.156	.060	.024	-.091	-.148
	.7428	.091	.055	-.083	-.156	.072	.024	-.063	-.124
	.7532	.099	.055	-.087	-.156	.072	.044	-.079	-.124
	.7806	.099	.052	-.083	-.128	.072	-.012	-.091	-.128
	.8628	.099	.026	-.127	-.144	.052	-.080	-.131	-.128
	.9313	.115	-.050	-.179	-.160	.000	-.104	-.179	-.172

TABLE 2

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

δ_f = 40 ° δ_a = 00 ° i_t = 00 °

$C_{\mu f}$ = .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 6.2 °				α = 9.8 °			
WING									
Upper	.0000	-5.194	-7.070	-1.450	-.853	-3.700	-1.616	-.867	-.596
	.0025	-3.736	-6.658	-1.434	-.782	-3.544	-2.122	-.892	-.592
	.0050	-3.317	-4.167	-1.360	-.782	-3.364	-1.518	-.846	-.592
	.0075	-2.933	-3.549	-1.360	-.778	-3.328	-1.569	-.846	-.592
	.0100	-2.597	-3.156	-1.360	-.778	-3.256	-1.565	-.842	-.592
	.0150	-2.060	-2.755	-1.380	-.790	-3.176	-1.541	-.859	-.588
	.0250	-1.474	-2.331	-1.368	-.782	-3.180	-1.584	-.842	-.576
	.0400	-1.150	-2.136	-1.391	-.738	-3.056	-1.549	-.842	-.572
	.0800	-.854	-1.588	-1.372	-.738	-2.504	-1.604	-.826	-.572
	.1300	-.589	-.879	-1.318	-.734	-.856	-1.498	-.792	-.572
	.1700	-.621	-.755	-1.333	-.706	-.736	-1.522	-.776	-.572
	.2800	-.569	-.580	-1.244	-.643	-.568	-1.341	-.726	-.568
	.4300	-.570	-.500	-.841	-.563	-.416	-1.150	-.668	-.560
	.5800	-.577	-.494	-.508	-.460	-.500	-.828	-.610	-.552
	.7300	-.600	-.342	-.283	-.373	-.550	-.702	-.573	-.512
Lower	.0025	-.846	-2.541	-.244	-.365	-1.480	-1.094	-.236	-.424
	.0050	-.138	-1.521	-.035	.135	-.524	-.659	-.046	.044
	.0100	.277	.043	.236	.325	.060	.055	.203	.252
	.0250	.459	.409	.415	.405	.452	.404	.398	.392
	.0400	.463	.412	.434	.361	.500	.420	.444	.356
	.0700	.407	.377	.399	.409	.504	.424	.419	.424
	.1700	.293	.280	.271	.175	.392	.333	.299	.184
	.2800	.277	.249	.198	.095	.356	.298	.224	.104
	.4300	.178	.257	.089	.008	.252	.275	.100	-.008
	.5800	.134	.257	.000	-.071	.152	.275	-.021	-.116
	.7200	.459	.339	-.058	-.143	.400	.357	-.112	-.208
FLAP or AILERON									
Upper	.7310	-.617	-.121	-.035	-.179	-.584	-.549	-.154	-.264
	.7395	-1.917	-.763	-.200	-.179	-1.892	-.965	-.191	-.244
	.7532	-1.941	-.514	-.163	-.294	-1.964	-.930	-.643	-.492
	.7806	-1.048	-.307	-.248	-.317	-1.076	-.631	-.585	-.492
	.8354	-.917	-.303	-.182	-.317	-.816	-.565	-.585	-.492
	.9038	-.609	-.303	-.159	-.298	-.420	-.475	-.556	-.464
	.9440	-.376	-.300	-.109	-.274	-.224	-.443	-.552	-.448
	.9863	-.300	-.290	-.093	-.278	-.150	-.420	-.510	-.424
	.9928	-.202	-.280	-.093	-.254	-.132	-.396	-.440	-.396
Lower	.7312	.510	.350	-.093	-.167	.410	.369	-.158	-.236
	.7367	.498	.350	-.093	-.167	.420	.345	-.129	-.240
	.7428	.451	.366	-.074	-.151	.444	.377	-.095	-.224
	.7532	.482	.362	-.070	-.151	.488	.380	-.129	-.224
	.7806	.565	.381	-.062	-.131	.544	.396	-.104	-.216
	.8628	.443	.179	-.062	-.131	.464	.200	-.170	-.216
	.9313	.316	.070	-.054	-.127	.308	.170	-.257	-.216

TABLE 2 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

δ_f = 40° δ_a = 00° i_t = 00°

$C_{\mu f}$ = .000 $C_{\mu,LE}$: I = .000 , C = .000 , O = .000 $C_{\mu,K}$: I = .000 , C = .000 , O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 13.7°				α = 17.5°			
WING									
Upper	.0000	-2.402	-.951	-.720	-.625	-1.848	-.992	-.732	-.643
	.0025	-2.299	-1.346	-.741	-.620	-1.725	-1.167	-.748	-.639
	.0050	-2.178	-.951	-.708	-.616	-1.688	-.941	-.724	-.639
	.0075	-2.195	-.946	-.708	-.616	-1.688	-.979	-.724	-.639
	.0100	-2.157	-.951	-.712	-.616	-1.688	-.962	-.724	-.631
	.0150	-2.170	-.934	-.712	-.616	-1.721	-.950	-.728	-.627
	.0250	-2.191	-.942	-.708	-.612	-1.697	-.962	-.732	-.627
	.0400	-2.236	-.938	-.708	-.596	-1.733	-.937	-.720	-.598
	.0800	-2.290	-.938	-.704	-.596	-1.565	-.941	-.720	-.602
	.1300	-2.050	-.938	-.700	-.596	-1.389	-.900	-.720	-.602
	.1700	-1.879	-.918	-.687	-.596	-1.467	-.900	-.720	-.602
	.2800	-1.456	-.881	-.683	-.592	-1.410	-.841	-.707	-.598
	.4300	-.860	-.850	-.663	-.616	-.856	-.830	-.691	-.598
	.5800	-.519	-.745	-.663	-.612	-.830	-.812	-.687	-.598
	.7300	-.450	-.724	-.642	-.580	.176	-.774	-.671	-.566
Lower	.0025	-1.411	-.807	-.300	-.494	-1.336	-.941	-.459	-.619
	.0050	-.622	-.560	-.132	-.082	-.656	-.682	-.276	-.221
	.0100	.004	.062	.140	.171	-.070	-.050	.000	.061
	.0250	.456	.420	.346	.339	.406	.356	.280	.287
	.0400	.531	.416	.407	.347	.500	.410	.370	.352
	.0700	.556	.412	.391	.437	.553	.448	.402	.422
	.1700	.461	.366	.309	.229	.500	.402	.337	.250
	.2800	.411	.309	.218	.131	.463	.351	.248	.172
	.4300	.295	.272	.103	.012	.311	.310	.126	.061
	.5800	.162	.235	-.025	-.094	.189	.264	.004	-.061
	.7200	.494	.300	-.148	-.225	.475	.314	-.134	-.180
FLAP or AILERON									
Upper	.7310	-.402	-.778	-.222	-.245	-.791	-.870	-.228	-.234
	.7395	-1.755	-.774	-.222	-.249	-1.004	-.866	-.240	-.234
	.7532	-2.299	-.700	-.642	-.584	-1.074	-.774	-.703	-.611
	.7806	-1.062	-.634	-.625	-.588	-.873	-.732	-.675	-.582
	.8354	-.647	-.654	-.630	-.571	-.762	-.715	-.683	-.541
	.9038	-.452	-.613	-.625	-.531	-.676	-.674	-.687	-.525
	.9440	-.373	-.580	-.617	-.531	-.619	-.665	-.687	-.525
	.9863	-.350	-.560	-.621	-.510	-.610	-.650	-.650	-.525
	.9928	-.332	-.551	-.621	-.514	-.582	-.628	-.691	-.525
Lower	.7312	.498	.300	-.132	-.233	.500	.351	-.122	-.193
	.7367	.498	.300	-.136	-.225	.471	.347	-.102	-.193
	.7428	.423	.337	-.078	-.200	.480	.364	-.077	-.172
	.7532	.502	.366	-.119	-.225	.496	.402	-.126	-.205
	.7806	.568	.337	-.078	-.188	.570	.402	-.077	-.168
	.8628	.448	.107	-.177	-.229	.418	.113	-.199	-.217
	.9313	.282	-.111	-.284	-.257	.217	-.130	-.297	-.270

TABLE 3

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} = .000$ $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = -1.2°				α = 6.1°			
WING									
Upper	.0000	.020	-.289	-.493	-.309	-5.108	-7.444	-3.012	-1.609
	.0025	-.479	-.937	-1.196	-.929	-4.945	-6.677	-2.837	-1.047
	.0150	-.266				-1.920			
	.0250	-.274	-.586	-.658	-.706	-1.447	-2.080	-2.745	-1.029
	.0400	-.209	-.557	-.526	-.454	-1.096	-1.648	-2.807	-1.008
	.0700	-.209	-.376	-.417	-.392	-.836	-1.249	-2.845	-.991
	.1200	-.221	-.376	-.349	-.367	-.665	-.982	-1.899	-1.000
	.1900	-.245	-.376	-.281	-.351	-.581	-.783	-.920	-.940
	.2200	-.245	-.351	-.349	-.318	-.527	-.720	-.591	-.853
	.2800	-.270	-.392	-.317	-.289	-.527	-.660	-.504	-.819
	.4300	-.306	-.367	-.269	-.231	-.468	-.559	-.395	-.708
	.5800	-.411	-.392	-.204	-.165	-.502	-.487	-.333	-.557
	.6800	-.540	-.446	-.200	-.152	-.569	-.487	-.283	-.484
Lower	.0025	.447	.409	.313	.322	-1.271	-2.118	-.574	-.296
	.0050	.423	.293	.457	.367	-.410	-.627	-.304	.137
	.0100	.334	.396	.437	.314	.092	-.050	.029	.343
	.0250	.181	.338	.333	.247	.305	.432	.416	.411
	.0700	.181	.082	.244	.115	.414	.584	.441	.326
	.1200	.129	.280	.212	.095	.372	.560	.429	.287
	.2200	.004	.235	.108	.016	.066	.402	.287	.163
	.2800	.173	.140	.124	-.008	.267	.266	.270	.141
	.4300	.161	.219	.028	-.090	.263	.317	.120	.004
	.5800	.374	.309	-.052	-.095	.426	.347	.012	-.068
	.7200	.475	.309	-.020	.000	.502	.372	-.016	-.017
FLAP or AILERON									
Upper	.7312	-.102	-.065			-.088	-.059		
	.7395	-.123	-.084			-.095	-.065		
	.7532	-.653	-.392	-.010	-.006	-.573	-.351	-.011	-.022
	.7806	-.616	-.392	-.208	-.119	-.560	-.338	-.249	-.369
	.8354	-.620	-.417	-.160	-.037	-.560	-.334	-.195	-.334
	.9038	-.689	-.400	-.120	-.012	-.627	-.334	-.154	-.283
	.9440	-.649	-.388	-.096	.004	-.585	-.334	-.145	-.283
	.9863	-.620	-.351	-.136	.000	-.020	-.241	-.124	-.257
	.9928	-.528	-.371	-.064	.000	-.456	-.326	-.079	-.214
Lower	.7312	.564	.330	.000	-.132	.615	.423	-.016	-.124
	.7367	.544	.330	.008	-.123	.594	.423	-.016	-.128
	.7428	.524	.347	.044	.049	.594	.466	.020	.038
	.7532	.524	.334	.012	-.099	.594	.415	-.012	-.128
	.7806	.520	.301	.040	-.045	.594	.389	.016	-.068
	.8628	.524	.227	.040	-.020	.556	.292	.016	-.077
	.9313	.338	.053	.004	.008	.376	.131	-.020	-.102

TABLE 3 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 9.7 °				α = 13.7 °			
WING									
Upper	.0000	-5.377	-1.619	-.946	-.713	-2.269	-.965	-.689	-.624
	.0025	-4.031	-1.606	-.946	-.629	-2.184	-.956	-.694	-.607
	.0150	-3.116				-2.089			
	.0250	-3.071	-1.584	-.920	-.612	-2.055	-.938	-.685	-.612
	.0400	-2.896	-1.589	-.889	-.599	-2.008	-.938	-.676	-.581
	.0700	-2.553	-1.554	-.889	-.599	-2.081	-.917	-.676	-.581
	.1200	-1.761	-1.523	-.849	-.595	-2.136	-.917	-.676	-.659
	.1900	-.968	-1.436	-.800	-.572	-1.879	-.856	-.458	-.568
	.2200	-.743	-1.384	-.791	-.572	-1.587	-.830	-.668	-.568
	.2800	-.684	-1.305	-.760	-.572	-1.398	-.799	-.668	-.564
	.4300	-.594	-1.043	-.681	-.559	-.811	-.765	-.654	-.573
	.5800	-.702	-.794	-.606	-.515	-.639	-.704	-.637	-.573
	.6800	-.927	-.663	-.561	-.515	-.682	-.704	-.637	-.564
Lower	.0025	-2.364	-1.139	-.181	-.246	-1.673	-.795	-.148	-.344
	.0050	-1.089	-.270	-.022	.101	-.858	-.165	-.039	.025
	.0100	-.225	.087	.150	.277	-.205	.121	.130	.224
	.0250	.207	.436	.407	.361	.158	.421	.375	.349
	.0700	.414	.659	.455	.303	.411	.660	.449	.318
	.1200	.427	.600	.451	.286	.476	.143	.449	.318
	.2200	.040	.488	.309	.162	.090	.482	.314	.193
	.2800	.355	.427	.305	.127	.407	.465	.314	.159
	.4300	.351	.375	.137	-.008	.394	.356	.117	.021
	.5800	.463	.371	.000	-.096	.484	.360	-.030	-.086
	.7200	.558	.371	-.057	-.088	.574	.360	-.109	-.064
FLAP or AILERON									
Upper	.7312	-.192	-.077			-.139	-.096		
	.7395	-.305	-.082			-.232	-.112		
	.7532	-1.571	-.567	-.057	-.055	-1.416	-.860	-.066	-.059
	.7806	-1.522	-.532	-.522	-.471	-1.077	-.760	-.659	-.543
	.8354	-1.400	-.532	-.513	-.475	-.832	-.673	-.619	-.499
	.9038	-.891	-.471	-.508	-.453	-.626	-.591	-.619	-.474
	.9440	-.612	-.436	-.490	-.418	-.523	-.539	-.318	-.422
	.9863	-.283	-.392	-.451	-.418	-.347	-.452	-.580	-.422
	.9928	-.373	-.397	-.455	-.418	-.463	-.504	-.598	-.443
Lower	.7312	.648	.401	-.084	-.198	.652	.443	-.078	-.185
	.7367	.639	.401	-.070	-.198	.647	.443	-.074	-.176
	.7428	.639	.436	-.039	-.026	.647	.495	-.017	-.012
	.7532	.639	.406	-.070	-.193	.647	.439	-.056	-.185
	.7806	.639	.358	-.070	-.132	.647	.408	-.056	-.133
	.8628	.572	.235	-.110	-.180	.557	.273	-.117	-.172
	.9313	.418	.052	-.199	-.229	.390	.108	-.240	-.215

TABLE 4

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^{\circ}$ $\delta_a = 00^{\circ}$ $i_t = 00^{\circ}$

$C_{\mu_f} = .014$ $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = -1.7°				α = 5.6°			
WING									
Upper	.0000	-.323	-1.143	-1.872	-1.195	-6.050	-7.981	-1.583	-1.172
	.0025	-.857	-1.677	-2.563	-1.524	-5.725	-6.233	-1.561	-.909
	.0150	-.441				-2.166			
	.0250	-.415	-.899	-1.143	-.916	-1.632	-2.886	-1.519	-.909
	.0400	-.336	-.817	-.885	-.666	-1.251	-2.499	-1.493	-.887
	.0700	-.310	-.569	-.711	-.558	-.957	-2.034	-1.458	-.887
	.1200	-.310	-.543	-.580	-.524	-.760	-1.538	-1.368	-1.400
	.1900	-.344	-.565	-.432	-.449	-.675	-1.095	-.922	-.874
	.2200	-.352	-.552	-.546	-.395	-.628	-.969	-1.197	-.870
	.2800	-.394	-.586	-.499	-.374	-.628	-.882	-1.145	-.840
	.4300	-.491	-.647	-.419	-.283	-.636	-.834	-.956	-.741
	.5800	-.747	-.860	-.330	-.208	-.828	-.960	-.759	-.616
	.6800	-1.176	-1.282	-.275	-.187	-1.217	-1.260	-.660	-.538
Lower	.0025	.394	.308	.148	.254	-1.666	-2.734	-.364	-.314
	.0050	.407	.308	.351	.379	-.628	-.886	-.175	.077
	.0100	.352	.482	.419	.379	.004	-.208	.042	.280
	.0250	.205	.434	.389	.312	.273	.378	.364	.366
	.0700	.226	.217	.292	.166	.410	.630	.424	.297
	.1200	.172	.273	.262	.137	.388	.121	.420	.267
	.2200	-.042	.312	.135	.029	.064	.447	.270	.150
	.2800	.176	.121	.131	.012	.307	.247	.270	.124
	.4300	.210	.282	-.004	-.062	.329	.352	.085	-.012
	.5800	.432	.330	-.122	-.095	.482	.360	-.055	-.107
	.7200	.495	.286	-.088	-.029	.546	.304	-.081	-.068
FLAP or AILERON									
Upper	.7312	-.266	-.297			-.254	-.262		
	.7395	-.632	-.618			-.594	-.500		
	.7532	-5.915	-6.185	-.058	-.039	-5.409	-4.846	-.126	-.097
	.7806	-1.852	-2.195	-.584	-.262	-1.700	-1.690	-.871	-.543
	.8354	-.743	-1.108	-.292	-.133	-.679	-.873	-.570	-.461
	.9038	-.264	-.734	-.182	-.054	-.213	-.569	-.459	-.396
	.9440	-.004	-.395	-.237	-.074	.042	-.360	-.351	-.353
	.9863	.200	-.250	-.200	-.050	.162	-.308	-.304	-.327
	.9928	.235	-.208	-.046	-.016	.243	-.234	-.308	-.327
Lower	.7312	.621	.360	-.114	-.145	.653	.395	-.047	-.163
	.7367	.600	.360	-.114	-.137	.636	.386	-.047	-.159
	.7428	.575	.378	-.072	-.029	.636	.439	.017	-.038
	.7532	.575	.352	-.097	-.129	.636	.378	-.030	-.185
	.7806	.575	.295	-.067	-.066	.636	.334	-.025	-.112
	.8628	.600	.204	-.050	-.045	.619	.208	-.060	-.146
	.9313	.516	.073	-.046	-.012	.521	.056	-.107	-.176

TABLE 4 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .014 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 9.6 °				α = 13.6 °			
WING									
Upper	.0000	-4.021	-1.669	-.890	-.839	-2.288	-.991	-.725	-.687
	.0025	-3.366	-1.651	-.890	-.728	-2.206	-.978	-.725	-.636
	.0150	-2.887				-2.120			
	.0250	-2.892	-1.634	-.872	-.742	-2.077	-.951	-.716	-.602
	.0400	-2.736	-1.608	-.872	-.715	-2.034	-.951	-.720	-.598
	.0700	-2.461	-1.560	-.859	-.715	-2.090	-.943	-.720	-.598
	.1200	-1.805	-1.508	-.855	-.879	-2.146	-.934	-.708	-.700
	.1900	-.978	-1.456	-.578	-.675	-1.900	-.886	-.484	-.568
	.2200	-.724	-1.456	-.784	-.693	-1.599	-.856	-.699	-.568
	.2800	-.637	-1.369	-.771	-.697	-1.405	-.830	-.699	-.568
	.4300	-.577	-1.108	-.754	-.733	-.840	-.791	-.699	-.568
	.5800	-.724	-.908	-.723	-.733	-.698	-.747	-.682	-.568
	.6800	-1.060	-.804	-.723	-.719	-.767	-.747	-.665	-.568
Lower	.0025	-2.064	-1.195	-.192	-.355	-1.680	-.891	-.205	-.363
	.0050	-.930	-.326	-.030	.031	-.862	-.186	-.072	.000
	.0100	-.185	.039	.131	.217	-.193	.113	.085	.205
	.0250	.224	.404	.381	.337	.185	.417	.356	.341
	.0700	.400	.656	.434	.293	.418	.665	.429	.329
	.1200	.435	.156	.425	.271	.487	.134	.429	.320
	.2200	.090	.473	.289	.146	.107	.482	.296	.196
	.2800	.374	.378	.258	.119	.426	.443	.291	.170
	.4300	.370	.360	.100	-.026	.409	.343	.124	.021
	.5800	.504	.352	-.070	-.119	.504	.343	-.047	-.076
	.7200	.573	.304	-.109	-.111	.581	.343	-.042	-.076
FLAP or AILERON									
Upper	.7312	-.228	-.128			-.132	-.103		
	.7395	-.561	-.171			-.377	-.149		
	.7532	-5.107	-1.751	-.099	-.093	-3.525	-1.586	-.100	-.088
	.7806	-1.525	-.904	-.762	-.715	-1.111	-.799	-.720	-.623
	.8354	-.603	-.669	-.662	-.657	-.719	-.695	-.656	-.568
	.9038	-.198	-.604	-.613	-.604	-.607	-.656	-.647	-.529
	.9440	.000	-.504	-.394	-.497	-.508	-.608	-.403	-.418
	.9863	.100	-.565	-.574	-.546	-.474	-.595	-.613	-.512
	.9928	.219	-.443	-.508	-.519	-.396	-.565	-.605	-.529
Lower	.7312	.672	.369	-.026	-.168	.659	.404	.055	-.119
	.7367	.659	.365	-.030	-.159	.655	.382	.055	-.102
	.7428	.659	.408	.026	-.004	.659	.452	.120	.089
	.7532	.659	.369	-.017	-.164	.659	.399	.072	-.102
	.7806	.659	.321	-.017	-.115	.668	.352	.072	-.064
	.8628	.607	.178	-.083	-.173	.577	.195	-.021	-.115
	.9313	.525	.008	-.201	-.244	.413	.013	-.175	-.205

TABLE 5

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

δ_f = 60 ° δ_a = 00 ° i_t = 00 °

C_{μ_f} = .031 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = -1.8 °				α = 5.5 °			
WING									
Upper	.0000	-.336	-1.204	-2.147	-1.219	-6.514	-4.627	-1.620	-1.149
	.0025	-.849	-1.695	-2.821	-1.723	-6.092	-3.994	-1.629	-.907
	.0150	-.431				-2.306			
	.0250	-.400	-.908	-1.227	-.982	-1.755	-3.163	-1.593	-.894
	.0400	-.314	-.817	-.973	-.727	-1.350	-2.910	-1.562	-.867
	.0700	-.284	-.582	-.781	-.618	-1.039	-2.530	-1.535	-.867
	.1200	-.293	-.552	-.638	-.591	-.844	-2.004	-1.446	-1.339
	.1900	-.327	-.573	-.473	-.477	-.759	-1.318	-.973	-.872
	.2200	-.353	-.552	-.589	-.429	-.706	-1.101	-1.254	-.872
	.2800	-.387	-.599	-.549	-.390	-.706	-.893	-1.196	-.858
	.4300	-.482	-.669	-.455	-.289	-.728	-.836	-1.022	-.797
	.5800	-.758	-.899	-.352	-.210	-.937	-1.004	-.825	-.687
	.6800	-1.219	-1.382	-.316	-.192	-1.373	-1.402	-.727	-.629
Lower	.0025	.405	.195	.102	.206	-1.875	-2.079	-.424	-.352
	.0050	.426	.286	.312	.363	-.755	-.614	-.227	.066
	.0100	.366	.465	.392	.381	-.057	-.088	.008	.264
	.0250	.249	.426	.370	.320	.239	.429	.343	.356
	.0700	.245	.221	.285	.188	.404	.632	.401	.290
	.1200	.181	.256	.254	.140	.377	.159	.388	.264
	.2200	-.012	.334	.129	.039	.031	.464	.254	.140
	.2800	.202	.208	.129	.008	.306	.283	.227	.118
	.4300	.224	.295	-.031	-.078	.324	.376	.049	-.017
	.5800	.448	.321	-.160	-.118	.479	.367	-.102	-.114
	.7200	.525	.234	-.129	-.030	.551	.322	-.120	-.088
FLAP or AILERON									
Upper	.7312	-.245	-.304			-.262	-.277		
	.7395	-.690	-.706			-.685	-.620		
	.7532	-6.714	-8.228	-.078	-.044	-6.652	-6.989	-.198	-.148
	.7806	-1.900	-3.129	-.825	-.232	-1.866	-2.592	-1.325	-.744
	.8354	-.745	-1.477	-.424	-.100	-.728	-1.194	-.772	-.555
	.9038	-.245	-1.199	-.276	-.057	-.244	-.867	-.620	-.471
	.9440	-.012	-.804	-.276	-.057	-.004	-.530	-.428	-.400
	.9863	.176	-.625	-.227	-.039	-.031	-.424	-.406	-.374
	.9928	.284	-.408	-.102	-.039	.284	-.221	-.348	-.352
Lower	.7312	.633	.295	-.183	-.175	.657	.389	-.066	-.171
	.7367	.620	.295	-.174	-.166	.639	.384	-.066	-.162
	.7428	.620	.339	-.124	-.092	.639	.433	.000	-.030
	.7532	.590	.282	-.165	-.149	.639	.367	-.058	-.158
	.7806	.594	.226	-.120	-.092	.644	.309	-.040	-.110
	.8628	.620	.056	-.120	-.061	.617	.159	-.080	-.140
	.9313	.555	-.086	-.107	-.026	.537	.000	-.156	-.171

TABLE 5 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^{\circ}$ $\delta_a = 00^{\circ}$ $i_t = 00^{\circ}$

$C_{\mu_f} =$ .031 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 9.6°				α = 13.6°			
WING									
Upper	.0000	-6.239	-1.625	-.862	-.858	-2.259	-.982	-.746	-.709
	.0025	-4.351	-1.603	-.849	-.734	-2.193	-.982	-.746	-.647
	.0150	-2.986				-2.087			
	.0250	-2.965	-1.585	-.844	-.716	-2.061	-.955	-.733	-.638
	.0400	-2.810	-1.558	-.844	-.712	-2.021	-.955	-.719	-.629
	.0700	-2.454	-1.526	-.831	-.712	-2.092	-.947	-.719	-.625
	.1200	-1.613	-1.450	-.823	-.889	-2.140	-.933	-.719	-.709
	.1900	-.849	-1.396	-.581	-.694	-1.911	-.885	-.488	-.572
	.2200	-.660	-1.427	-.793	-.703	-1.612	-.881	-.719	-.599
	.2800	-.643	-1.342	-.767	-.703	-1.422	-.850	-.706	-.599
	.4300	-.617	-1.103	-.767	-.729	-.867	-.797	-.706	-.599
	.5800	-.789	-.914	-.758	-.729	-.722	-.744	-.706	-.599
	.6800	-1.145	-.837	-.754	-.729	-.779	-.744	-.693	-.577
Lower	.0025	-2.445	-1.153	-.206	-.309	-1.678	-.828	-.186	-.334
	.0050	-1.158	-.337	-.021	.039	-.850	-.171	-.079	-.008
	.0100	-.266	.018	.137	.234	-.198	.110	.097	.193
	.0250	.180	.382	.387	.345	.185	.422	.346	.352
	.0700	.429	.630	.431	.291	.418	.678	.413	.330
	.1200	.429	.112	.405	.269	.497	.132	.439	.330
	.2200	.068	.463	.275	.132	.123	.502	.311	.193
	.2800	.373	.423	.267	.115	.431	.431	.297	.154
	.4300	.373	.328	.086	-.030	.422	.370	.106	.022
	.5800	.480	.319	-.077	-.123	.510	.365	-.044	-.092
	.7200	.566	.279	-.077	-.110	.594	.334	-.111	-.092
FLAP or AILERON									
Upper	.7312	-.225	-.117			-.127	-.094		
	.7395	-.627	-.207			-.425	-.175		
	.7532	-6.093	-2.441	-.138	-.125	-4.109	-2.101	-.098	-.087
	.7806	-1.617	-1.180	-.900	-.805	-1.154	-1.057	-.799	-.678
	.8354	-.626	-.738	-.650	-.668	-.731	-.718	-.666	-.594
	.9038	-.223	-.648	-.573	-.614	-.616	-.691	-.657	-.572
	.9440	-.072	-.553	-.374	-.486	-.488	-.651	-.359	-.400
	.9863	.000	-.630	-.543	-.544	-.488	-.651	-.671	-.550
	.9928	.188	-.495	-.474	-.517	-.325	-.603	-.644	-.590
Lower	.7312	.643	.337	.060	-.137	.678	.383	-.066	-.145
	.7367	.622	.337	.060	-.119	.665	.378	-.062	-.145
	.7428	.622	.369	.120	.088	.665	.405	-.017	-.013
	.7532	.622	.342	.077	-.106	.665	.374	-.057	-.145
	.7806	.635	.283	.073	-.079	.665	.325	-.044	-.114
	.8628	.574	.144	-.021	-.132	.594	.162	-.128	-.176
	.9313	.489	-.036	-.150	-.216	.431	-.017	-.248	-.246

TABLE 6

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

δ_f = 60 ° δ_a = 00 ° i_t = 00 °

C_{μ_f} = .099 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α =-2.0 °				α = 5.3 °			
WING									
Upper	.0000	-.557	-1.594	-2.625	-2.025	-7.561	-3.238	-1.438	-1.174
	.0025	-1.085	-2.034	-3.263	-2.149	-6.927	-3.154	-1.443	-1.062
	.0150	-.570				-2.635			
	.0250	-.498	-1.013	-1.362	-1.154	-2.036	-3.039	-1.394	-1.062
	.0400	-.400	-.897	-1.065	-.867	-1.566	-3.004	-1.364	-1.045
	.0700	-.374	-.632	-.866	-.709	-1.192	-2.845	-1.351	-1.049
	.1200	-.374	-.620	-.698	-.590	-.968	-2.491	-1.285	-1.060
	.1900	-.408	-.632	-.500	-.526	-.872	-1.770	-1.230	-1.071
	.2200	-.417	-.632	-.629	-.474	-.826	-1.473	-1.166	-1.076
	.2800	-.464	-.667	-.590	-.449	-.836	-1.106	-1.123	-1.076
	.4300	-.579	-.752	-.496	-.329	-.863	-.960	-1.039	-1.040
	.5800	-.915	-1.030	-.392	-.256	-1.151	-1.159	-.947	-.911
	.6800	-1.485	-1.628	-.336	-.222	-1.694	-1.624	-.886	-.777
Lower	.0025	.362	.226	.022	.107	-2.320	-1.929	-.381	-.496
	.0050	.400	.252	.233	.329	-1.005	-.593	-.224	-.027
	.0100	.374	.466	.345	.376	-.169	-.119	.018	.205
	.0250	.255	.453	.375	.325	.210	.425	.329	.321
	.0700	.272	.282	.280	.184	.402	.703	.399	.268
	.1200	.200	.192	.254	.132	.379	.600	.395	.241
	.2200	.200	.355	.125	.021	.350	.522	.250	.112
	.2800	.196	.300	.112	-.026	.320	.450	.228	.067
	.4300	.251	.303	-.052	-.111	.356	.407	.039	-.062
	.5800	.464	.308	-.203	-.154	.534	.385	-.123	-.161
	.7200	.536	.226	-.177	-.081	.594	.283	-.184	-.129
FLAP or AILERON									
Upper	.7312	-.073	-.079			-.078	-.079		
	.7395	-1.301	-1.074			-1.352	-.943		
	.7532	-11.772	-15.395	-.100	-.073	-11.874	-13.628	-.144	-.111
	.7806	-2.566	-6.666	-1.422	-.679	-2.667	-5.601	-1.579	-.790
	.8354	-1.264	-2.632	-.547	-.252	-1.288	-1.929	-.886	-.598
	.9038	-.464	-1.521	-.276	-.115	-.466	-1.194	-.697	-.482
	.9440	-.400	-1.218	-.319	-.145	-.402	-1.119	-.622	-.393
	.9863	.511	-1.137	-.323	-.145	-.174	-1.099	-.579	-.384
	.9928	-.047	-.808	-.142	-.098	-.027	-.805	-.465	-.388
Lower	.7312	.634	.303	-.241	-.231	.689	.376	-.237	-.246
	.7367	.634	.278	-.224	-.205	.685	.349	-.206	-.237
	.7428	.613	.312	-.198	-.205	.685	.389	-.193	-.237
	.7532	.587	.256	-.228	-.205	.685	.336	-.215	-.241
	.7806	.613	.192	-.185	-.141	.685	.292	-.184	-.170
	.8628	.643	-.013	-.168	-.124	.662	.093	-.210	-.201
	.9313	.579	-.214	-.151	-.085	.603	-.128	-.272	-.205

TABLE 6 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

C_{μ_f} = .099 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 9.3 °				α = 13.3 °			
WING									
Upper	.0000	-5.449	-1.498	-1.036	-.851	-2.804	-1.119	-.819	-.773
	.0025	-4.097	-1.493	-1.036	-.799	-2.660	-1.119	-.819	-.729
	.0150	-3.246				-2.387			
	.0250	-3.211	-1.462	-1.018	-.782	-2.360	-1.093	-.811	-.712
	.0400	-3.097	-1.466	-1.000	-.768	-2.330	-1.084	-.811	-.690
	.0700	-2.828	-1.408	-1.000	-.773	-2.434	-1.079	-.811	-.694
	.1200	-2.026	-1.368	-.978	-.763	-2.434	-1.071	-.811	-.803
	.1900	-1.000	-1.287	-.943	-.773	-2.091	-1.004	-.533	-.694
	.2200	-.705	-1.238	-.937	-.795	-1.734	-.987	-.811	-.694
	.2800	-.670	-1.184	-.942	-.795	-1.478	-.960	-.819	-.699
	.4300	-.709	-1.045	-.910	-.816	-.861	-.902	-.828	-.738
	.5800	-.951	-.897	-.910	-.834	-.769	-.867	-.802	-.773
	.6800	-1.432	-.834	-.910	-.834	-.913	-.889	-.780	-.747
Lower	.0025	-2.564	-1.184	-.309	-.485	-2.052	-1.031	-.269	-.537
	.0050	-1.211	-.291	-.179	-.039	-1.087	-.283	-.181	-.096
	.0100	-.313	.054	.022	.179	-.313	.013	.009	.144
	.0250	.159	.457	.336	.310	.083	.389	.317	.310
	.0700	.423	.717	.417	.275	.391	.690	.410	.310
	.1200	.441	.600	.417	.258	.469	.600	.441	.306
	.2200	.420	.534	.296	.144	.450	.513	.330	.192
	.2800	.396	.453	.265	.109	.443	.478	.304	.157
	.4300	.396	.404	.076	-.044	.443	.385	.128	-.004
	.5800	.529	.395	-.099	-.140	.539	.363	-.044	-.109
	.7200	.630	.327	-.202	-.144	.600	.319	-.132	-.109
FLAP or AILERON									
Upper	.7312	-.068	-.062			-.070	-.063		
	.7395	-1.260	-.384			-.950	-.427		
	.7532	-11.281	-7.327	-.163	-.123	-8.359	-7.118	-.200	-.147
	.7806	-2.365	-3.193	-1.516	-.904	-1.608	-2.893	-1.568	-.882
	.8354	-1.092	-1.403	-.964	-.764	-.982	-1.332	-.916	-.694
	.9038	-.352	-.682	-.762	-.668	-.730	-.730	-.718	-.624
	.9440	-.335	-.592	-.600	-.572	-.617	-.699	-.650	-.546
	.9863	-.300	-.592	-.691	-.572	-.483	-.668	-.731	-.607
	.9928	.018	-.516	-.686	-.572	-.139	-.549	-.736	-.633
Lower	.7312	.692	.395	-.197	-.258	.674	.420	-.075	-.223
	.7367	.692	.381	-.175	-.231	.665	.389	-.079	-.214
	.7428	.692	.413	-.161	-.223	.661	.442	-.035	-.179
	.7532	.692	.359	-.206	-.240	.661	.380	-.075	-.223
	.7806	.705	.323	-.175	-.188	.674	.349	-.066	-.188
	.8628	.648	.161	-.256	-.236	.609	.195	-.150	-.231
	.9313	.586	-.009	-.350	-.293	.496	.027	-.295	-.297

TABLE 7

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} = .102$ $C_{\mu,LE}$: I = .004 ,C = .007 ,O = .016 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = -2.1°				α = 5.2°			
WING									
Upper	.0000	-.784	-2.098	-3.332	-2.497	-7.597	-14.518	-11.497	-13.326
	.0025	-.662	-2.615	-4.103	-2.567	-6.346	-10.453	-10.256	-12.606
	.0150	-.593				-2.515			
	.0250	-.519	-1.295	-1.409	-.720	-1.909	-3.547	-4.127	-3.477
	.0400	-.364	-1.047	-1.090	-.244	-1.355	-2.660	-2.963	-2.153
	.0700	-.442	-.602	-.715	-.380	-1.074	-1.495	-2.391	-1.859
	.1200	-.377	-.611	-.685	-.707	-.900	-1.369	-2.009	-1.938
	.1900	-.377	-.585	-.522	-.498	-.762	-1.078	-1.502	-1.502
	.2200	-.420	-.585	-.625	-.550	-.732	-1.052	-1.364	-1.057
	.2800	-.472	-.684	-.565	-.384	-.758	-1.052	-1.164	-.842
	.4300	-.571	-.748	-.465	-.323	-.766	-.974	-.823	-.631
	.5800	-.931	-1.055	-.328	-.244	-1.065	-1.230	-.614	-.469
	.6800	-1.506	-1.641	-.336	-.266	-1.602	-1.891	-.504	-.465
Lower	.0025	.290	-.013	-.086	.070	-2.294	-5.121	-2.886	-5.109
	.0050	.368	.218	.177	.323	-1.017	-1.865	-2.250	-2.280
	.0100	.368	.449	.336	.380	-.169	-.978	-1.223	-.978
	.0250	.251	.444	.405	.323	.229	.178	.086	.083
	.0700	.268	.303	.323	.183	.403	.669	.382	.311
	.1200	.190	.261	.293	.144	.364	.580	.418	.311
	.2200	.170	.363	.134	.022	.300	.509	.250	.189
	.2800	.186	.340	.129	-.013	.294	.450	.227	.149
	.4300	.238	.320	-.039	-.109	.355	.400	.009	.004
	.5800	.468	.320	-.198	-.162	.519	.365	-.168	-.096
	.7200	.537	.209	-.168	-.048	.584	.239	-.136	-.070
FLAP or AILERON									
Upper	.7312	-.025	-.032			-.028	-.046		
	.7395	-1.296	-1.186			-1.309	-1.242		
	.7532	-12.006	-16.451	-.151	-.102	-12.049	-15.737	-.217	-.154
	.7806	-2.623	-6.264	-1.776	-.961	-2.580	-5.908	-1.977	-1.092
	.8354	-1.294	-2.602	-.698	-.332	-1.216	-2.508	-.818	-.487
	.9038	-.502	-1.491	-.319	-.135	-.528	-1.491	-.520	-.380
	.9440	-.437	-1.261	-.323	-.170	-.407	-1.360	-.523	-.329
	.9863	-.300	-1.218	-.431	-.210	-.286	-1.252	-.455	-.303
	.9928	-.065	-.808	-.181	-.118	-.030	-.748	-.182	-.180
Lower	.7312	.615	.308	-.194	-.236	.667	.352	-.095	-.171
	.7367	.615	.278	-.172	-.205	.667	.326	-.082	-.153
	.7428	.584	.320	-.147	-.122	.667	.365	-.027	-.101
	.7532	.580	.256	-.172	-.196	.667	.317	-.077	-.153
	.7806	.606	.205	-.129	-.131	.667	.265	-.032	-.083
	.8628	.632	.017	-.129	-.105	.641	.065	-.045	-.105
	.9313	.563	-.175	-.129	-.096	.580	-.130	-.086	-.110

TABLE 7 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: Basic Wing

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .102 $C_{\mu,LE}$: I = .004 ,C = .007 ,O = .016 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 8.8 °				α = °			
WING									
Upper	.0000	-13.962	-3.567	-2.079	-3.843				
	.0025	-11.298	-3.644	-2.079	-5.262				
	.0150	-3.887							
	.0250	-2.892	-3.702	-2.219	-4.230				
	.0400	-2.081	-3.756	-2.251	-2.788				
	.0700	-1.513	-2.513	-2.121	-1.894				
	.1200	-1.207	-2.284	-1.688	-1.779				
	.1900	-1.009	-1.716	-1.523	-1.120				
	.2200	-.941	-1.630	-1.539	-1.083				
	.2800	-.941	-1.590	-1.521	-1.051				
	.4300	-.919	-1.441	-1.363	-.995				
	.5800	-1.180	-1.320	-1.135	-.899				
	.6800	-1.684	-1.441	-1.019	-.834				
Lower	.0025	-4.986	-2.973	-.958	-3.608				
	.0050	-2.567	-1.144	-.819	-1.760				
	.0100	-.968	-.658	-.460	-.834				
	.0250	-.234	.171	.172	.018				
	.0700	.302	.689	.386	.276				
	.1200	.410	.610	.437	.313				
	.2200	.410	.540	.302	.207				
	.2800	.419	.482	.288	.175				
	.4300	.419	.414	.098	.018				
	.5800	.568	.387	-.074	-.092				
	.7200	.631	.275	-.177	-.124				
FLAP or AILERON									
Upper	.7312	-.027	-.017						
	.7395	-1.337	-.857						
	.7532	-12.390	-11.418	-.260	-.213				
	.7806	-2.594	-3.846	-1.972	-1.290				
	.8354	-1.225	-1.387	-1.033	-.843				
	.9038	-.432	-.797	-.786	-.687				
	.9440	-.378	-.644	-.690	-.530				
	.9863	-.347	-.604	-.633	-.599				
	.9928	.009	-.405	-.549	-.604				
Lower	.7312	.721	.482	-.116	-.198				
	.7367	.716	.464	-.098	-.184				
	.7428	.730	.531	-.060	-.138				
	.7532	.725	.450	-.107	-.198				
	.7806	.721	.410	-.107	-.138				
	.8628	.676	.311	-.135	-.217				
	.9313	.622	.113	-.233	-.253				

TABLE 8

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat

δ_f = 40 ° δ_a = 00 ° i_t = 00°

C_{μ_f} = .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 6.1 °				α = 9.8 °			
WING									
Upper	.0000	-4.811	-8.230	-2.326	-4.471	-3.897	-8.633	-3.554	-5.709
	.0025	-3.517	-6.629	-2.066	-3.904	-3.430	-5.699	-3.269	-5.950
	.0050	-3.151	-5.060	-1.981	-2.712	-3.323	-5.950	-2.904	-3.709
	.0075	-2.795	-4.020	-1.946	-2.448	-3.250	-4.765	-2.759	-3.240
	.0100	-2.475	-3.396	-1.915	-2.256	-3.213	-4.119	-2.675	-2.896
	.0150	-2.019	-2.791	-1.868	-2.034	-3.188	-3.407	-2.474	-2.535
	.0250	-1.432	-2.052	-1.767	-1.732	-3.254	-2.518	-2.301	-2.265
	.0400	-1.081	-1.640	-1.647	-1.398	-3.213	-1.967	-2.064	-1.896
	.0800	-.815	-1.095	-1.450	-.877	-2.291	-1.288	-1.687	-1.191
	.1300		-.889	-1.151	-.701	-.852	-1.045	-1.333	-.987
	.1700	-.564	-.842	-.965	-.621	-.787	-.934	-1.120	-.875
	.2800	-.544	-.672	-.663	-.483	-.619	-.761	-.815	-.693
	.4300	-.520	-.550	-.485	-.360	-.484	-.670	-.598	-.527
	.5800	-.500	-.498	-.333	-.280	-.561	-.514	-.430	-.411
	.7300	-.500	-.379	-.233	-.157	-.450	-.346	-.305	-.253
Lower	.0025	-.776	-2.514	-.260	-1.532	-1.549	-2.453	-.454	-1.593
	.0050	-.062	-1.467	.012	-.115	-.590	-1.539	-.149	.008
	.0100	.309	.174	.306	.391	.090	.346	.237	.220
	.0250	.463	.585	.519	.498	.443	.646	.498	.506
	.0400	.425	.534	.519	.582	.500	.650	.538	.622
	.0700	.386	.480	.481	.448	.488	.621	.526	.564
	.1700	.290	.320	.283	.241	.393	.436	.361	.261
	.2800	.255	.237	.186	.080	.336	.325	.265	.087
	.4300	.154	.186	.097	-.038	.213	.284	.157	.008
	.5800	.100	.182	.008	-.115	.160	.251	.052	-.050
	.7200	.413	.316	-.070	-.119	.479	.350	-.028	-.104
FLAP or AILERON									
Upper	.7310	-.525	-.186	-.031	-.096	-.414	-.206	-.012	-.104
	.7395	-1.587	-.751	-.100	-.110	-1.692	-.588	-.250	-.180
	.7532	-1.313	-.581	-.213	-.146	-1.746	-.457	-.269	-.253
	.7806	-.795	-.352	-.221	-.146	-.885	-.296	-.265	-.220
	.8354	-.699	-.336	-.163	-.103	-.561	-.255	-.217	-.178
	.9038	-.544	-.328	-.140	-.077	-.348	-.226	-.177	-.149
	.9440	-.425	-.324	-.093	-.073	-.283	-.226	-.141	-.149
	.9863	-.400	-.300	-.078	-.050	-.230	-.225	-.116	-.120
	.9928	-.367	-.289	-.062		-.229	-.222	-.116	-.124
Lower	.7312	.483	.340	-.093	-.176	.496	.366	-.072	-.120
	.7367	.467	.340	-.070	-.146	.500	.370	-.056	-.100
	.7428	.413	.360	-.062	-.126	.434	.387	-.048	-.095
	.7532	.444	.360	-.062	-.130	.479	.416	-.048	-.095
	.7806	.525	.340	-.062	-.088	.549	.387	-.048	-.075
	.8628	.425	.142	-.062	-.046	.451	.210	-.056	-.075
	.9313	.255	.028	-.047	-.015	.283	.086	-.056	-.054
SLAT									
Upper	-.0200		-.900	-.996	-.924		-1.500	-1.354	-1.017
	-.0700		-.602	-.438	-.406		-1.008	-.775	-.646
	-.1250		-.367	-.223	-.100		-1.000	-.713	-.417
	-.1400		-.422	-.036	.000		-1.642	-1.004	-.508
	-.1450		-.382	.012	.167		-1.871	-.025	-.379
	-.1500		.283	.307	.319		-.996	-.233	.158

TABLE 8 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat

$\delta_f = 40^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu f}$ = .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 13.5°				α = 17.3°			
WING									
Upper	.0000	-2.055	-8.062	-3.296	-5.008	-1.913	-7.340	-2.512	-1.591
	.0025	-1.992	-5.572	-3.025	-5.305	-1.830	-5.614	-2.343	-1.418
	.0050	-1.958	-6.213	-2.779	-3.471	-1.813	-5.860	-2.049	-1.160
	.0075	-1.958	-4.984	-2.700	-3.045	-1.830	-4.954	-1.942	-1.122
	.0100	-1.954	-4.229	-2.638	-2.723	-1.821	-4.405	-1.884	-1.067
	.0150	-1.954	-3.641	-2.488	-2.397	-1.846	-3.713	-1.793	-1.067
	.0250	-1.954	-2.751	-2.354	-2.190	-1.846	-2.971	-1.661	-.983
	.0400	-1.962	-2.180	-2.163	-1.901	-1.884	-2.524	-1.545	-.827
	.0800	-1.950	-1.433	-1.892	-1.227	-1.767	-1.709	-1.368	-.802
	.1300	-1.840	-1.167	-1.604	-.938	-1.700	-1.262	-1.173	-.776
	.1700	-1.727	-1.016	-1.404	-.839	-1.693	-1.131	-1.025	-.738
	.2800	-1.458	-.751	-1.025	-.678	-1.585	-.885	-.802	-.650
	.4300	-.803	-.550	-.704	-.529	-1.017	-.740	-.686	-.591
	.5800	-.609	-.506	-.508	-.405	-.830	-.631	-.616	-.548
	.7300	-.500	-.433	-.404	-.355	-.650	-.496	-.583	-.506
Lower	.0025	-1.298	-1.861	-.463	-1.417	-1.568	-2.406	-.186	-.055
	.0050	-.462	-1.433	-.163	-.033	-.809	-1.955	.041	.367
	.0100	.004	.392	.208	.405	-.170	.238	.326	.494
	.0250	.433	.674	.483	.500	.357	.484	.529	.544
	.0400	.513	.700	.554	.583	.473	.600	.607	.580
	.0700	.534	.620	.554	.607	.544	.639	.624	.500
	.1700	.454	.482	.413	.306	.514	.545	.479	.194
	.2800	.416	.371	.292	.174	.477	.422	.343	.135
	.4300	.298	.302	.179	.079	.344	.361	.207	.097
	.5800	.164	.286	.054	.000	.187	.324	.087	-.017
	.7200	.300	.347	-.038	-.083	.510	.369	-.045	-.139
FLAP or AILERON									
Upper	.7310	-.479	-.404	-.033	-.124	-.693	-.516	-.004	-.160
	.7395	-1.198	-.465	-.033	-.300	-.875	-.516	-.300	-.400
	.7532	-1.542	-.412	-.379	-.331	-.963	-.459	-.578	-.536
	.7806	-.782	-.363	-.358	-.318	-.709	-.406	-.558	-.510
	.8354	-.508	-.343	-.317	-.306	-.564	-.373	-.562	-.506
	.9038	-.349	-.294	-.279	-.277	-.436	-.332	-.517	-.477
	.9440	-.261	-.294	-.254	-.277	-.365	-.303	-.479	-.473
	.9863	-.270	-.275	-.254	-.277	-.350	-.290	-.479	-.468
	.9928	-.223	-.269	-.242	-.273	-.324	-.275	-.459	-.468
Lower	.7312	.350	.339	-.071	-.120	.510	.402	-.066	-.160
	.7367	.370	.335	-.071	-.120	.510	.398	-.066	-.156
	.7428	.400	.359	-.038	-.116	.436	.406	-.066	-.139
	.7532	.471	.392	-.063	-.116	.510	.451	-.066	-.160
	.7806	.555	.363	-.067	-.116	.573	.410	-.066	-.148
	.8628	.445	.200	-.079	-.116	.461	.209	-.132	-.177
	.9313	.269	.045	-.108	-.124	.286	.053	-.202	-.228
SLAT									
Upper	-.0200		-1.700	-1.584	-1.139		-1.600	-1.815	-1.379
	-.0700		-1.376	-1.065	-.861		-1.617	-1.480	-1.172
	-.1250		-1.751	-1.314	-.763		-2.506	-2.194	-1.330
	-.1400		-3.070	-2.363	-1.082		-4.436	-4.423	-1.965
	-.1450		-4.180	.033	-1.086		-6.652	-5.000	-2.022
	-.1500		-3.796	-2.163	-.841		-7.607	-5.630	-2.744

TABLE 9

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat

δ_f= 60 ° δ_a= 00 ° i_t= 00 °

$C_{\mu f}$= .041 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.5 °				α = 9.3 °			
WING									
Upper	.0000	-6.786	-7.470	-4.280	-6.016	-3.359	-9.285	-4.395	-6.557
	.0025	-4.451	-6.733	-3.979	-6.254	-3.256	-6.277	-4.047	-6.893
	.0050	-3.895	-6.491	-3.329	-3.914	-3.182	-6.883	-3.513	-4.523
	.0075	-3.399	-5.168	-3.160	-3.414	-3.165	-5.510	-3.370	-3.961
	.0100	-2.972	-4.471	-3.037	-3.045	-3.144	-4.809	-3.273	-3.510
	.0150	-2.423	-3.733	-2.839	-2.668	-3.095	-4.070	-3.084	-3.055
	.0250	-1.754	-2.750	-2.629	-2.344	-3.095	-3.066	-2.920	-2.732
	.0400	-1.452	-2.184	-2.378	-1.893	-3.033	-2.444	-2.656	-2.327
	.0800	-1.040	-1.467	-1.872	-1.197	-2.847	-1.651	-2.248	-1.506
	.1300	-.689	-1.233	-1.416	-1.037	-1.500	-1.394	-1.849	-1.204
	.1700	-.726	-1.176	-1.226	-.926	-1.112	-1.265	-1.576	-1.072
	.2800	-.726	-1.049	-.983	-.738	-.760	-1.046	-1.185	-.864
	.4300	-.700	-1.100	-.786	-.578	-.700	-1.000	-.874	-.685
	.5800	-.818	-1.127	-.572	-.426	-.748	-.958	-.643	-.532
	.7300	-.880	-1.373	-.424	-.299	-1.050	-1.066	-.517	-.408
Lower	.0025	-1.496	-2.442	-.663	-1.787	-1.607	-2.419	-.744	-1.957
	.0050	-.464	-1.758	-.313	-.082	-.640	-1.983	-.382	-.213
	.0100	.149	.238	.136	.455	.021	.324	.092	.379
	.0250	.460	.607	.453	.550	.455	.550	.437	.500
	.0400	.492	.650	.535	.590	.533	.600	.534	.583
	.0700	.456	.631	.510	.545	.541	.589	.559	.587
	.1700	.371	.414	.346	.209	.463	.502	.412	.230
	.2800	.343	.369	.222	.025	.434	.398	.273	.106
	.4300	.290	.348	.078	-.066	.364	.386	.143	.034
	.5800	.387	.332	-.086	-.135	.413	.365	.008	-.064
	.7200	.609	.283	-.177	-.176	.624	.353	-.105	-.123
FLAP or AILERON									
Upper	.7310	-5.552	-5.565	-.321	-.217	-4.822	-3.800	-.172	-.179
	.7395	-6.927	-7.811	-.444	-.213	-5.979	-4.722	-.219	-.191
	.7532	-6.641	-7.549	-.634	-.377	-5.929	-4.327	-.567	-.443
	.7806	-1.903	-2.504	-.564	-.348	-1.603	-1.265	-.513	-.417
	.8354	-.762	-1.377	-.362	-.254	-.665	-.631	-.387	-.323
	.9038	-.270	-1.152	-.300	-.184	-.306	-.581	-.332	-.281
	.9440	-.093	-.807	-.226	-.176	-.132	-.465	-.277	-.255
	.9863	.200	-.557	-.210	-.152	.000	-.328	-.269	-.238
	.9928	.226	-.459	-.206	-.164	.169	-.282	-.252	-.217
Lower	.7312	.508	.332	-.321	-.250	.578	.407	-.219	-.187
	.7367	.395	.295	-.272	-.217	.517	.357	-.181	-.166
	.7428	.399	.357	-.235	-.217	.434	.361	-.164	-.149
	.7532	.698	.316	-.263	-.221	.550	.336	-.164	-.149
	.7806	.738	.258	-.210	-.180	.748	.295	-.164	-.149
	.8628	.661	.061	-.226	-.160	.657	.178	-.164	-.128
	.9313	.573	-.094	-.198	-.139	.554	.066	-.164	-.132
SLAT									
Upper	-.0200		-1.450	-1.392	-1.106		-2.150	-1.691	-1.197
	-.0700		-1.053	-.690	-.588		-1.473	-1.045	-.811
	-.1250		-.935	-.559	-.290		-1.679	-1.132	-.638
	-.1400		-1.412	-.608	-.294		-2.864	-1.852	-.823
	-.1450		-1.604	-.751	-.106		-3.741	-1.675	-.753
	-.1500		-.678	.102	.367		-3.111	-1.370	-.292

TABLE 9 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .041 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 13.2°				α = 17.1°			
WING									
Upper	.0000	-2.654	-7.812	-4.071	-4.962	-2.141	-7.565	-2.211	-1.417
	.0025	-2.485	-6.030	-3.754	-5.180	-2.064	-5.932	-2.077	-1.298
	.0050	-2.304	-6.400	-3.171	-3.556	-2.030	-6.240	-1.772	-1.051
	.0075	-2.296	-5.189	-3.025	-3.197	-2.034	-5.160	-1.699	-.991
	.0100	-2.243	-4.601	-2.959	-2.962	-2.034	-4.586	-1.646	-.962
	.0150	-2.206	-3.992	-2.771	-2.701	-2.064	-3.911	-1.577	-.940
	.0250	-2.210	-2.983	-2.629	-2.569	-2.064	-3.219	-1.512	-.894
	.0400	-2.271	-2.446	-2.429	-2.325	-2.103	-2.852	-1.439	-.800
	.0800	-2.148	-1.689	-2.109	-1.787	-2.005	-1.945	-1.325	-.774
	.1300	-1.860	-1.391	-1.746	-1.363	-1.950	-1.477	-1.102	-.745
	.1700	-1.749	-1.256	-1.529	-1.180	-1.910	-1.308	-.894	-.715
	.2800	-1.638	-.966	-1.167	-.953	-1.821	-1.055	-.829	-.677
	.4300	-.850	-.850	-.871	-.774	-1.090	-.950	-.768	-.655
	.5800	-.732	-.740	-.638	-.641	-.966	-.814	-.699	-.634
	.7300	-.900	-.727	-.546	-.543	-1.100	-.764	-.663	-.608
Lower	.0025	-1.613	-2.277	-.663	-1.376	-1.816	-2.557	-.114	-.051
	.0050	-.774	-1.836	-.342	-.128	-1.004	-1.915	.061	.319
	.0100	-.091	.269	.108	.321	-.282	.063	.301	.400
	.0250	.391	.487	.442	.453	.299	.342	.524	.549
	.0400	.506	.500	.525	.509	.449	.460	.610	.620
	.0700	.551	.559	.583	.633	.530	.646	.626	.685
	.1700	.514	.504	.433	.274	.556	.561	.467	.153
	.2800	.469	.408	.283	.141	.509	.451	.321	.123
	.4300	.395	.403	.150	.038	.427	.405	.191	.081
	.5800	.436	.374	.046	-.043	.453	.380	.061	-.043
	.7200	.613	.374	-.075	-.124	.603	.359	-.069	-.149
FLAP or AILERON									
Upper	.7310	-3.395	-2.416	-.150	-.248	-3.197	-2.287	-.114	-.221
	.7395	-4.099	-2.416	-.175	-.248	-3.530	-2.198	-.122	-.234
	.7532	-4.222	-2.240	-.571	-.551	-3.428	-2.063	-.687	-.647
	.7806	-1.177	-.773	-.500	-.509	-1.098	-.810	-.638	-.596
	.8354	-.613	-.521	-.429	-.474	-.688	-.574	-.618	-.574
	.9038	-.354	-.429	-.375	-.449	-.427	-.485	-.593	-.549
	.9440	-.230	-.382	-.333	-.432	-.291	-.430	-.577	-.545
	.9863	-.100	-.340	-.304	-.415	-.200	-.400	-.573	-.545
	.9928	.000	-.277	-.300	-.406	-.056	-.359	-.585	-.545
Lower	.7312	.605	.408	-.121	-.184	.598	.418	-.175	-.208
	.7367	.584	.403	-.121	-.175	.564	.422	-.142	-.213
	.7428	.506	.408	-.075	-.154	.500	.409	-.118	-.183
	.7532	.469	.382	-.092	-.162	.483	.375	-.142	-.200
	.7806	.687	.361	-.079	-.162	.701	.346	-.138	-.200
	.8628	.597	.219	-.117	-.162	.581	.211	-.199	-.230
	.9313	.502	.139	-.158	-.218	.479	.093	-.248	-.289
SLAT									
Upper	-.0200		-1.950	-1.808	-1.265		-1.800	-1.924	-1.428
	-.0700		-1.674	-1.249	-1.000		-1.597	-1.563	-1.237
	-.1250		-2.290	-1.690	-.996		-2.627	-2.385	-1.449
	-.1400		-3.968	-3.143	-1.376		-4.390	-4.991	-2.152
	-.1450		-5.772	-2.772	-1.457		-6.783	-4.021	-2.326
	-.1500		-5.976	-3.560	-1.498		-8.389	-6.872	-3.339

TABLE 10

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu f}$ = .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of – 0.31, α = -1.1°	0.56, α = -1.1°	0.76, α = -1.1°	0.90, α = -1.1°	0.31, α = 6.1°	0.56, α = 6.1°	0.76, α = 6.1°	0.90, α = 6.1°
WING									
Upper	.0000	-.553	-.570	-1.599	-1.210	-.839	-.828	-2.182	-2.102
	.0025	-.500	-.824	-1.424	-1.027	-1.250	-1.408	-2.938	-3.315
	.0050	-.530	-.660	-1.039	-.805	-1.600	-1.572	-2.690	-2.596
	.0075	-.395	-.609	-.899	-.744	-1.727	-1.496	-2.508	-2.278
	.0100	-.391	-.570	-.852	-.699	-1.924	-1.424	-2.066	-2.065
	.0150	-.395	-.523	-.735	-.634	-1.707	-1.308	-1.785	-1.755
	.0250	-.328	-.496	-.615	-.561	-1.253	-1.216	-1.459	-1.465
	.0400	-.229	-.453	-.525	-.477	-.952	-1.036	-1.194	-1.180
	.0800	-.233	-.414	-.436	-.389	-.759	-.868	-.930	-.894
	.1300	-.237	-.367	-.405	-.363	-.546	-.704	-.835	-.825
	.1700	-.241	-.371	-.389	-.340	-.546	-.704	-.744	-.743
	.2800	-.281	-.387	-.354	-.309	-.502	-.612	-.649	-.584
	.4300	-.300	-.380	-.319	-.256	-.273	-.510	-.525	-.465
	.5800	-.372	-.422	-.268	-.198	-.470	-.504	-.376	-.343
	.7300	-.430	-.250	-.183	-.099	-.420	-.450	-.260	-.212
Lower	.0025	-.130	-.098	-.459	-.790	-.349	.220	.062	-.053
	.0050	-.043	-.062	-.288	-.435	-.150	.352	.256	.433
	.0100	.150	.074	-.058	-.191	.096	.416	.380	.543
	.0250	.270	.262	.191	.038	.420	.388	.450	.522
	.0400	.270	.301	.311	.120	.520	.348	.492	.478
	.0700	.210	.203	.315	.168	.510	.368	.459	.380
	.1700	.119	.129	.202	.095	.293	.280	.326	.159
	.2800	.119	.156	.093	.053	.293	.236	.211	.102
	.4300	.095	.215	.031	-.008	.205	.284	.124	.004
	.5800	.047	.309	-.016	-.053	.112	.316	.012	-.082
	.7200	.419	.355	-.047	-.080	.462	.380	-.070	-.122
FLAP or AILERON									
Upper	.7310	-.775	-.777	-.459	-.141	-.743	-.724	-.388	-.200
	.7395	-.727	-.738	-.284	-.076	-.675	-.612	-.178	-.127
	.7532	-.632	-.445	-.222	-.076	-.578	-.384	-.211	-.147
	.7806	-.585	-.445	-.300	-.156	-.566	-.384	-.306	-.229
	.8354	-.589	-.445	-.249	-.065	-.594	-.384	-.252	-.159
	.9038	-.636	-.445	-.230	-.042	-.635	-.384	-.219	-.118
	.9440	-.573	-.441	-.183	-.034	-.558	-.388	-.169	-.110
	.9863	-.550	-.430	-.175	-.030	-.520	-.370	-.150	-.082
	.9928	-.478	-.410	-.167	-.027	-.446	-.364	-.145	-.082
Lower	.7312	.569	.395	.012	-.050	.590	.430	-.012	-.082
	.7367	.573	.398	.012	-.050	.590	.436	-.021	-.078
	.7428	.561	.410	.047	-.034	.590	.436	.012	-.078
	.7532	.518	.410	.004	-.065	.518	.408	-.045	-.078
	.7806	.549	.363	.023	-.034	.590	.380	-.004	-.078
	.8628	.522	.219	-.035	-.011	.510	.236	-.066	-.078
	.9313	.293	.025	-.093	-.008	.309	.152	-.033	-.037
SLAT									
Upper	-.0200		-.198	-.422	-.240		-.457	-1.082	-.967
	-.0700		-.298	-.202	-.442		-1.110	-1.000	-1.057
	-.1250		.147	.217	.291		-.706	-.559	-.302
	-.1400		.291	.345	.380		-.902	-.563	-.241
	-.1450		.023	.302	.384		-.857	-.747	-.041
	-.1500		-.267	-.500	-1.233		-.233	.049	.367

TABLE 10 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 9.8°				α = 13.4°			
WING									
Upper	.0000	-.375	-.870	-2.758	-2.313	-1.000	-1.250	-3.751	-2.366
	.0025	-.930	-1.610	-4.024	-4.123	-2.250	-2.000	-5.331	-4.642
	.0050	-2.100	-2.004	-3.587	-3.369	-3.500	-2.317	-4.670	-3.870
	.0075	-2.785	-1.935	-3.222	-3.004	-4.136	-2.374	-4.115	-3.512
	.0100	-2.711	-1.833	-2.885	-2.766	-3.706	-2.292	-3.743	-3.256
	.0150	-2.410	-1.710	-2.496	-2.369	-3.409	-2.189	-3.257	-2.821
	.0250	-1.824	-1.607	-1.988	-1.996	-2.521	-2.053	-2.576	-2.362
	.0400	-1.367	-1.365	-1.595	-1.603	-1.930	-1.753	-2.086	-1.915
	.0800	-1.023	-1.135	-1.226	-1.230	-1.409	-1.432	-1.584	-1.496
	.1300	-.691	-.905	-1.083	-1.103	-.905	-1.136	-1.359	-1.321
	.1700	-.695	-.865	-.972	-.968	-.880	-1.053	-1.192	-1.171
	.2800	-.594	-.742	-.782	-.770	-.727	-.868	-.951	-.923
	.4300	-.450	.131	-.623	-.595	-.650	-.680	-.706	-.695
	.5800	-.469	-.528	-.429	-.433	-.529	-.584	-.486	-.524
	.7300	-.430	-.500	-.321	-.302	-.450	-.440	-.355	-.366
Lower	.0025	-.258	.190	.067	.040	-.140	.050	-.057	-.028
	.0050	-.050	.286	.254	.150	.033	.200	.176	.187
	.0100	.100	.306	.429	.270	.145	.320	.412	.320
	.0250	.380	.350	.488	.397	.320	.430	.539	.530
	.0400	.450	.350	.524	.470	.420	.480	.567	.600
	.0700	.488	.298	.524	.500	.470	.500	.588	.565
	.1700	.410	.321	.413	.159	.500	.400	.453	.211
	.2800	.387	.325	.282	.095	.467	.374	.327	.183
	.4300	.301	.325	.159	.056	.384	.354	.208	.126
	.5800	.156	.341	.044	-.008	.198	.374	.110	.041
	.7200	.500	.409	-.040	-.067	.574	.416	-.004	-.033
FLAP or AILERON									
Upper	.7310	-.660	-.659	-.226	-.159	-.686	-.601	-.159	-.118
	.7395	-.570	-.536	-.075	-.131	-.603	-.481	-.020	-.085
	.7532	-.523	-.345	-.250	-.230	-.512	-.333	-.261	-.276
	.7806	-.508	-.262	-.294	-.274	-.508	-.321	-.290	-.297
	.8354	-.551	-.254	-.250	-.218	-.558	-.325	-.229	-.285
	.9038	-.566	-.325	-.190	-.175	-.566	-.321	-.192	-.232
	.9440	-.504	-.317	-.147	-.151	-.475	-.317	-.131	-.183
	.9863	-.450	-.300	-.119	-.123	-.450	-.315	-.139	-.163
	.9928	-.391	-.298	-.107	-.103	-.388	-.300	-.098	-.163
Lower	.7312	.617	.470	-.004	-.044	.632	.480	.069	.004
	.7367	.613	.452	-.008	-.024	.636	.494	.041	.004
	.7428	.605	.452	.020	-.016	.640	.461	.061	.004
	.7532	.574	.448	-.024	-.056	.599	.449	.024	-.020
	.7806	.637	.413	-.020	-.040	.657	.399	.037	-.016
	.8628	.543	.274	-.067	-.067	.591	.263	-.016	-.028
	.9313	.359	.143	-.016	-.016	.421	.230	.033	-.028
SLAT									
Upper	-.0200		-.667	-1.541	-1.626		-.858	-1.900	-2.243
	-.0700		-1.866	-1.703	-1.553		-2.707	-2.301	-1.958
	-.1250		-1.695	-1.362	-.919		-2.987	-2.414	-1.674
	-.1400		-2.906	-2.248	-1.089		-4.950	-4.644	-2.230
	-.1450		-2.250	-2.159	-.959		-4.000	-4.088	-2.255
	-.1500		-2.935	-1.992	-.390		-7.920	-5.971	-2.423

TABLE 11

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

δ_f = 60° δ_a = 00° i_t = 00°

$C_{\mu f}$ = .027 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = -1.7°				α = 5.5°			
WING									
Upper	.0000	-.300	-.832	-2.181	-1.720	-.386	-.827	-2.672	-2.304
	.0025	-.500	-.850	-2.079	-1.594	-.500	-1.450	-3.820	-3.901
	.0050	-.600	-.884	-1.547	-1.167	-1.000	-1.904	-3.412	-3.185
	.0075	-.618	-.800	-1.244	-1.071	-2.313	-1.827	-3.084	-2.811
	.0100	-.578	-.760	-1.177	-1.013	-2.329	-1.751	-2.736	-2.560
	.0150	-.566	-.720	-1.039	-.895	-2.024	-1.635	-2.356	-2.197
	.0250	-.482	-.696	-.843	-.795	-1.528	-1.542	-1.916	-1.848
	.0400	-.335	-.640	-.709	-.649	-1.175	-1.321	-1.560	-1.494
	.0800	-.331	-.588	-.583	-.519	-.923	-1.133	-1.232	-1.140
	.1300	-.259	-.512	-.551	-.510	-.626	-.964	-1.100	-1.016
	.1700	-.319	-.552	-.524	-.460	-.675	-.964	-1.012	-.934
	.2800	-.350	-.600	-.496	-.402	-.675	-.892	-.852	-.749
	.4300	-.335	-.700	-.480	-.343	-.650	-.950	-.708	-.613
	.5800	-.580	-.904	-.450	-.272	-.732	-1.088	-.520	-.457
	.7300	-.700	-.950	-.450	-.272	-.850	-1.300	-.400	-.272
Lower	.0025	.300	-.108	-.394	-.925	.200	.185	.100	.400
	.0050	.327	.112	-.154	-.364	.358	.345	.232	.470
	.0100	.350	.216	.130	-.059	.420	.361	.400	.550
	.0250	.402	.264	.382	.192	.550	.450	.492	.580
	.0400	.351	.272	.429	.360	.570	.450	.492	.600
	.0700	.351	.336	.382	.515	.541	.382	.492	.564
	.1700	.163	.272	.217	.075	.366	.317	.340	.270
	.2800	.187	.244	.091	.025	.366	.313	.224	.074
	.4300	.171	.276	-.008	-.050	.309	.341	.068	-.012
	.5800	.271	.316	-.122	-.117	.329	.341	-.068	-.086
	.7200	.586	.304	-.193	-.126	.618	.321	-.148	-.119
FLAP or AILERON									
Upper	.7310	-4.163	-4.120	-.362	-.126	-4.280	-4.321	-.528	-.284
	.7395	-6.705	-7.764	-.394	-.126	-6.695	-7.984	-.304	-.193
	.7532	-6.777	-7.984	-.450	-.126	-6.618	-8.076	-.356	-.288
	.7806	-1.892	-2.572	-.402	-.209	-1.805	-2.558	-.472	-.358
	.8354	-.801	-1.352	-.205	-.084	-.732	-1.325	-.292	-.218
	.9038	-.311	-1.172	-.157	-.071	-.260	-1.161	-.224	-.156
	.9440	-.112	-.852	-.102	-.071	-.098	-.803	-.136	-.140
	.9863	.378	-.640	-.094	-.038	.200	-.582	-.124	-.111
	.9928	.187	-.492	-.063	-.038	.276	-.402	-.088	-.086
Lower	.7312	.669	.304	-.268	-.142	.691	.350	-.164	-.123
	.7367	.673	.376	-.236	-.134	.699	.398	-.168	-.119
	.7428	.677	.360	-.232	-.134	.699	.378	-.152	-.119
	.7532	.629	.336	-.232	-.134	.654	.349	-.152	-.123
	.7806	.629	.260	-.232	-.121	.695	.261	-.152	-.119
	.8628	.665	.068	-.189	-.067	.667	.096	-.152	-.091
	.9313	.550	-.072	-.150	.000	.589	-.048	-.130	-.070
SLAT									
Upper	-.0200		-.300	-.578	-.700		-.950	-1.376	-1.550
	-.0700		-.414	-.349	-.542		-1.649	-1.408	-1.339
	-.1250		.072	.153	.265		-1.331	-.902	-.551
	-.1400		.229	.337	.378		-2.159	-1.347	-.571
	-.1450		.096	.233	.414		-.314	-1.404	-.376
	-.1500		.016	-.201	-.940		-1.743	-.788	.196

TABLE 11 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

$\delta_f = 60^{\circ}$ $\delta_a = 00^{\circ}$ $i_t = 00^{\circ}$

$C_{\mu f}$ = .027 $C_{\mu,LE}$: I = .000 ,C =.000 ,O =.000 $C_{\mu,K}$: I = .000 ,C =.000 ,O =.000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 9.1°				α = 12.8°			
WING									
Upper	.0000	-.079	-.831	-3.693	-2.323	-1.087	-1.200	-4.610	-2.894
	.0025	-1.000	-2.087	-5.236	-4.749	-1.200	-2.429	-6.406	-5.368
	.0050	-2.222	-2.339	-4.630	-3.872	-4.206	-2.664	-5.538	-4.525
	.0075	-3.577	-2.289	-4.087	-3.498	-4.661	-2.698	-4.885	-4.101
	.0100	-3.159	-2.227	-3.718	-3.281	-4.000	-2.651	-4.525	-3.818
	.0150	-2.908	-2.095	-3.249	-2.783	-3.669	-2.551	-3.970	-3.296
	.0250	-2.176	-1.983	-2.601	-2.396	-2.698	-2.450	-3.174	-2.741
	.0400	-1.661	-1.707	-2.116	-1.953	-2.103	-2.130	-2.576	-2.275
	.0800	-1.255	-1.475	-1.660	-1.536	-1.545	-1.752	-1.979	-1.775
	.1300	-.849	-1.207	-1.436	-1.374	-1.190	-1.433	-1.703	-1.568
	.1700	-.845	-1.153	-1.286	-1.230	-1.008	-1.357	-1.534	-1.407
	.2800	-.778	-1.062	-1.070	-.979	-.839	-1.193	-1.220	-1.148
	.4300	-.730	-1.065	-.871	-.766	-.700	-1.190	-.987	-.945
	.5800	-.766	-1.066	-.639	-.604	-.731	-1.200	-.678	-.720
	.7300	-.850	-1.100	-.570	-.413	-.850	-1.250	-.564	-.559
Lower	.0025	.200	.150	.150	.250	.100	.200	.200	.150
	.0050	.250	.200	.250	.311	.150	.250	.250	.200
	.0100	.300	.250	.382	.196	.250	.310	.320	.300
	.0250	.400	.350	.506	.106	.450	.400	.466	.400
	.0400	.450	.400	.544	.072	.480	.450	.551	.420
	.0700	.448	.400	.544	.070	.500	.450	.564	.400
	.1700	.485	.347	.382	.111	.533	.420	.411	.203
	.2800	.444	.388	.241	.072	.517	.399	.271	.191
	.4300	.385	.388	.095	-.009	.442	.399	.161	.097
	.5800	.385	.364	-.025	-.085	.405	.366	.047	-.004
	.7200	.649	.302	-.120	-.123	.661	.350	-.068	-.085
FLAP or AILERON									
Upper	.7310	-3.983	-4.334	-.597	-.217	-3.405	-4.206	-.589	-.580
	.7395	-6.339	-8.020	-.550	-.204	-5.355	-7.576	-.381	-.600
	.7532	-6.431	-8.057	-.456	-.383	-5.355	-7.572	-.564	-.517
	.7806	-1.715	-2.533	-.531	-.383	-1.314	-2.315	-.623	-.530
	.8354	-.640	-1.335	-.340	-.298	-.517	-1.172	-.381	-.411
	.9038	-.192	-1.219	-.257	-.238	-.161	-1.084	-.288	-.352
	.9440	-.033	-.893	-.183	-.226	-.050	-.761	-.250	-.305
	.9863	.150	-.694	-.145	-.200	.170	-.576	-.161	-.288
	.9928	.272	-.488	-.145	-.162	.285	-.437	-.161	-.258
Lower	.7312	.720	.350	-.145	-.081	.727	.370	-.076	-.042
	.7367	.724	.376	-.158	-.094	.727	.382	-.076	-.034
	.7428	.724	.364	-.124	-.098	.727	.382	-.076	-.038
	.7532	.669	.326	-.124	-.106	.686	.332	-.076	-.038
	.7806	.724	.223	-.124	-.111	.735	.248	-.076	-.038
	.8628	.682	.054	-.124	-.098	.694	.084	-.076	-.038
	.9313	.590	-.091	-.054	-.094	.628	-.067	-.076	-.051
SLAT									
Upper	-.0200		-1.500	-1.815	-1.872		-1.950	-2.103	-2.420
	-.0700		-2.416	-2.037	-1.728		-3.185	-2.588	-2.090
	-.1250		-2.432	-1.881	-1.132		-3.741	-2.848	-1.905
	-.1400		-3.959	-3.370	-1.494		-6.234	-5.765	-2.609
	-.1450		-4.500	-3.400	-1.457		-1.165	-4.868	-2.667
	-.1500		-5.596	-3.757	-1.140		-11.143	-8.226	-3.436

TABLE 12

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

δ_f= 50 ° δ_a= 50 ° i_t= 00 °

$C_{\mu f}$= .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = -1.4 °				α = 5.8 °			
WING									
Upper	.0000	-.050	-.421	-1.804	-1.685	-.183	-.682	-3.453	-3.226
	.0025	-.150	-.516	-1.487	-1.348	-.350	-1.107	-3.945	-3.799
	.0050	-.200	-.563	-1.117	-1.135	-.750	-1.615	-3.711	-2.894
	.0075	-.312	-.532	-.993	-1.026	-1.470	-1.575	-2.586	-2.676
	.0100	-.301	-.508	-.917	-.947	-1.155	-1.476	-2.476	-2.518
	.0150	-.406	-.488	-.759	-.824	-1.633	-1.377	-2.137	-2.115
	.0250	-.327	-.468	-.638	-.712	-1.219	-1.254	-1.644	-1.755
	.0400	-.252	-.413	-.536	-.569	-.924	-1.032	-1.293	-1.395
	.0800	-.252	-.381	-.438	-.468	-.741	-.869	-.988	-1.071
	.1300	-.237	-.349	-.411	-.442	-.558	-.730	-.867	-.976
	.1700	-.256	-.373	-.396	-.408	-.546	-.702	-.793	-.862
	.2800	-.301	-.389	-.381	-.378	-.478	-.639	-.711	-.719
	.4300	-.300	-.300	-.381	-.367	-.500	-.550	-.605	-.621
	.5800	-.398	-.350	-.300	-.345	-.530	-.540	-.449	-.546
	.7300	-.550	-.400	.136	-.350	-.600	-.540	-.450	-.470
Lower	.0025	.038	-.091	-.355	-.288	.239	.202	-.160	-.142
	.0050	.150	-.008	.300	-.461	.275	.313	.066	.115
	.0100	.230	.079	-.162	-.109	.275	.385	.316	.316
	.0250	.330	.258	.170	.150	.331	.397	.437	.427
	.0400	.330	.298	.294	.322	.323	.377	.473	.455
	.0700	.280	.250	.389	.404	.335	.333	.461	.478
	.1700	.079	.095	.294	.240	.303	.230	.367	.202
	.2800	.090	.143	.223	.206	.271	.242	.301	.154
	.4300	.060	.226	.245	.184	.211	.270	.270	.182
	.5800	.030	.286	.287	.236	.108	.317	.297	.182
	.7200	.342	.405	.404	.311	.339	.413	.383	.241
FLAP or AILERON									
Upper	.7310	-.771	-.746	-.566	-.652	-.685	-.746	-.652	-.723
	.7395	-1.271	-.996	-.577	-.491	-1.044	-.948	-.637	-.613
	.7532	-.748	-.552	-.389	-.371	-.693	-.536	-.410	-.431
	.7806	-.722	-.524	-.362	-.330	-.653	-.512	-.402	-.423
	.8354	-.699	-.516	-.366	-.367	-.653	-.532	-.426	-.463
	.9038	-.744	-.480	-.351	-.371	-.645	-.552	-.426	-.482
	.9440	-.620	-.480	-.359	-.360	-.506	-.512	-.426	-.518
	.9863	-.600	-.478	-.343	-.360	-.475	-.490	-.414	-.514
	.9928	-.534	-.476	-.359	-.367	-.456	-.472	-.437	-.506
Lower	.7312	.477	.420	.408	.307	.514	.508	.418	.273
	.7367	.455	.452	.404	.322	.498	.508	.375	.265
	.7428	.447	.448	.396	.322	.482	.480	.402	.300
	.7532	.398	.421	.396	.322	.462	.476	.434	.304
	.7806	.474	.417	.408	.367	-.500	.456	.422	.348
	.8628	.432	.306	.309	.292	.454	.361	.289	.213
	.9313	.229	.130	.257	.165	.295	.282	.160	.000
SLAT									
Upper	-.0200		-.159	-.363	-.159		-.700	-1.200	-1.000
	-.0700		-.196	-.170	-.052		-1.073	-1.019	-.850
	-.1250		.230	.211	.307		-.690	-.601	-.356
	-.1400		.296	.278	.363		-.935	-.728	-.333
	-.1450		.274	.278	.330		-1.057	-.873	-.153
	-.1500		-.452	-.696	-1.463		-.287	-.100	.295

TABLE 12 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

$\delta_f = 50^{\circ}$ $\delta_a = 50^{\circ}$ $i_t = 00^{\circ}$

$C_{\mu f}$ = .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 9.4°				α = 13.1°			
WING									
Upper	.0000	-.400	-.839	-4.547	-4.204	-.850	-1.388	-6.036	-5.727
	.0025	-1.250	-1.831	-5.325	-5.197	-2.200	-2.075	-6.908	-6.967
	.0050	-2.500	-2.084	-4.623	-3.738	-3.250	-2.459	-5.422	-4.339
	.0075	-2.839	-2.000	-3.547	-3.393	-3.598	-2.399	-4.618	-4.000
	.0100	-2.610	-1.916	-3.361	-3.165	-3.676	-2.328	-4.229	-3.735
	.0150	-2.542	-1.807	-2.897	-2.738	-3.348	-2.182	-3.643	-3.273
	.0250	-1.900	-1.659	-2.202	-2.247	-2.451	-2.020	-2.855	-2.657
	.0400	-1.450	-1.394	-1.754	-1.828	-1.856	-1.727	-2.245	-2.182
	.0800	-1.076	-1.149	-1.333	-1.377	-1.361	-1.360	-1.651	-1.640
	.1300	-.751	-.932	-1.123	-1.232	-.922	-1.099	-1.382	-1.450
	.1700	-.727	-.900	-1.008	-1.075	-.852	-1.024	-1.189	-1.269
	.2800	-.626	-.783	-.821	-.859	-.701	-.862	-.948	-1.000
	.4300	-.550	-.650	-.659	-.718	-.640	.150	-.691	-.744
	.5800	-.558	-.590	-.524	-.604	-.598	-.569	-.486	-.595
	.7300	-.600	-.550	-.500	-.500	-.450	-.500	-.400	-.470
Lower	.0025	.193	.056	-.175	-.176	.111	-.364	-.341	-.306
	.0050	.225	.165	-.020	-.133	.200	-.170	-.209	-.150
	.0100	.300	.300	.298	.133	.330	.020	.221	.012
	.0250	.420	.450	.476	.267	.420	.260	.450	-.320
	.0400	.450	.480	.496	.400	.450	.360	.498	-.400
	.0700	.420	.450	.571	.533	.480	.420	.566	-.450
	.1700	.410	.253	.456	.231	.488	.308	.458	.248
	.2800	.382	.281	.357	.255	.447	.360	.361	.252
	.4300	.301	.325	.313	.220	.361	.368	.329	.252
	.5800	.145	.345	.306	.239	.197	.368	.325	.227
	.7200	.502	.442	.369	.251	.434	.435	.365	.211
FLAP or AILERON									
Upper	.7310	-.795	-.687	-.607	-.647	-.787	-.565	-.534	-.607
	.7395	-1.317	-.791	-.500	-.502	-1.488	-.629	-.438	-.488
	.7532	-.835	-.466	-.365	-.384	-.795	-.348	-.305	-.372
	.7806	-.739	-.430	-.345	-.384	-.734	-.344	-.305	-.376
	.8354	-.735	-.478	-.381	-.404	-.709	-.344	-.305	-.413
	.9038	-.759	-.494	-.377	-.439	-.635	-.348	-.305	-.438
	.9440	-.655	-.458	-.381	-.463	-.550	-.340	-.305	-.450
	.9863	-.640	-.450	-.369	-.463	-.500	-.330	-.305	-.450
	.9928	-.598	-.434	-.365	-.459	-.410	-.324	-.305	-.446
Lower	.7312	.578	.450	.385	.271	.615	.500	.426	.256
	.7367	.570	.502	.377	.267	.627	.538	.426	.264
	.7428	.558	.466	.377	.286	.590	.502	.426	.269
	.7532	.560	.458	.417	.271	.570	.502	.426	.273
	.7806	.590	.462	.321	.290	.570	.498	.426	.273
	.8628	.518	.313	.302	.208	.553	.360	.361	.211
	.9313	.329	.100	.254	.145	.389	.289	.293	.128
SLAT									
Upper	-.0200		-1.450	-1.800	-1.569		-1.750	-2.350	-2.180
	-.0700		-1.740	-1.577	-1.000		-2.496	-2.204	-1.650
	-.1250		-1.573	-1.309	-.886		-2.792	-2.320	-1.604
	-.1400		-2.671	-2.163	-1.049		-4.560	-4.500	-2.240
	-.1450		-3.171	-2.069	-.935		-6.500	-3.984	-2.160
	-.1500		-2.630	-1.959	-.423		-7.204	-5.908	-2.520

TABLE 13

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

$\delta_f = 50^\circ$ $\delta_a = 50^\circ$ $i_t = 00^\circ$

$C_{\mu_f} = .027$ $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = -2.3°				α = 4.9°			
WING									
Upper	.0000	-.250	-.685	-3.004	-2.743	-.600	-.861	-4.741	-4.056
	.0025	-.300	-.750	-2.912	-2.606	-.800	-1.422	-5.546	-4.992
	.0050	-.400	-.829	-2.032	-2.293	-1.000	-2.000	-4.817	-3.916
	.0075	-.530	-.769	-1.680	-1.944	-2.276	-1.906	-3.741	-3.728
	.0100	-.455	-.737	-1.568	-1.819	-1.798	-1.803	-3.479	-3.492
	.0150	-.526	-.693	-1.288	-1.550	-1.897	-1.721	-3.055	-3.020
	.0250	-.407	-.669	-1.084	-1.269	-1.444	-1.565	-2.356	-2.584
	.0400	-.300	-.618	-.904	-1.060	-1.095	-1.340	-1.898	-2.156
	.0800	-.296	-.538	-.748	-.867	-.860	-1.135	-1.487	-1.652
	.1300	-.241	-.486	-.712	-.831	-.621	-.984	-1.301	-1.544
	.1700	-.293	-.542	-.708	-.799	-.654	-.979	-1.212	-1.400
	.2800	-.383	-.574	-.736	-.779	-.638	-.914	-1.119	-1.248
	.4300	-.450	-.650	-.796	-.863	-.560	-.950	-1.093	-1.244
	.5800	-.542	-.857	-.820	-1.020	-.687	-1.049	-1.034	-1.400
	.7300	-.700	-1.150	-.900	-1.250	-.900	-1.250	-1.200	-1.600
Lower	.0025	.178	-.080	-.252	-.137	.247	.119	-.220	-.140
	.0050	.206	.072	-.060	-.008	.296	.200	-.013	.000
	.0100	.300	.163	.236	.329	.370	.320	.292	.200
	.0250	.430	.283	.420	.454	.500	.450	.470	.420
	.0400	.450	.339	.456	.466	.480	.480	.500	.500
	.0700	.430	.303	.460	.534	.450	.324	.551	.500
	.1700	.158	.247	.344	.265	.358	.287	.453	.196
	.2800	.166	.263	.308	.205	.358	.328	.369	.232
	.4300	.154	.323	.308	.237	.296	.381	.352	.184
	.5800	.277	.402	.348	.273	.366	.418	.369	.164
	.7200	.494	.478	.448	.297	.556	.484	.394	.104
FLAP or AILERON									
Upper	.7310	.330	.500	.500	-.800	.350	.390	.400	-.470
	.7395	-4.495	-4.789	-4.416	-4.321	-4.629	-5.016	-4.720	-4.704
	.7532	-5.103	-5.582	-5.520	-5.630	-5.127	-5.762	-5.716	-6.132
	.7806	-1.510	-1.721	-1.964	-2.337	-1.448	-1.709	-2.017	-2.880
	.8354	-.644	-.693	-.840	-1.008	-.638	-.647	-.830	-1.240
	.9038	-.221	-.410	-.416	-.502	-.185	-.455	-.398	-.648
	.9440	-.051	-.355	-.180	-.293	-.033	-.418	-.144	-.424
	.9863	.100	-.195	-.044	-.177	.150	-.234	-.021	-.296
	.9928	.198	-.155	.040	-.092	.247	-.242	.072	-.204
Lower	.7312	.565	.550	.480	.357	.642	.500	.449	.140
	.7367	.565	.550	.460	.337	.646	.578	.449	.144
	.7428	.569	.522	.488	.337	.650	.557	.449	.148
	.7532	.522	.526	.488	.340	.601	.557	.453	.088
	.7806	.565	.502	.488	.365	.654	.533	.475	.148
	.8628	.597	.462	.432	.305	.654	.492	.436	.120
	.9313	.466	.406	.350	.200	.551	.410	.400	.080
SLAT									
Upper	-.0200		-.450	-.571	-.273		-1.200	-1.800	-1.229
	-.0700		-.469	-.490	.094		-1.652	-1.352	-.650
	-.1250		-.024	-.073	.253		-1.407	-.889	-.486
	-.1400		.151	.102	.355		-2.249	-1.269	-.518
	-.1450		.253	.539	.363		-2.530	-1.301	-.340
	-.1500		.086	-.155	-.886		-1.973	-.723	.182

TABLE 13 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON, and SLAT

Wing leading edge configuration: 0.60b/2 Slat + Rad.

δ_f = 50 ° δ_a = 50 ° i_t = 00 °

$C_{\mu f}$ = .027 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of –							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 8.6 °				α = 12.3 °			
WING									
Upper	.0000	-2.611	-1.293	-6.209	-5.617	-.500	-1.783	-8.176	-7.448
	.0025	-3.750	-1.980	-7.168	-6.907	-.900	-2.396	-9.100	-8.866
	.0050	-4.500	-2.406	-5.596	-4.742	-1.500	-2.904	-6.624	-5.184
	.0075	-5.058	-2.329	-4.809	-4.520	-3.866	-2.821	-6.058	-4.925
	.0100	-2.817	-2.248	-4.470	-4.242	-3.946	-2.725	-5.603	-4.648
	.0150	-3.213	-2.114	-3.845	-3.677	-3.656	-2.571	-4.873	-3.996
	.0250	-2.100	-1.972	-2.992	-3.133	-2.681	-2.409	-3.801	-3.381
	.0400	-1.550	-1.683	-2.421	-2.597	-2.051	-2.071	-2.932	-2.807
	.0800	-1.150	-1.386	-1.833	-2.004	-1.496	-1.654	-2.185	-2.255
	.1300	-.850	-1.159	-1.584	-1.855	-.979	-1.379	-1.873	-2.046
	.1700	-.800	-1.138	-1.470	-1.685	-.979	-1.338	-1.713	-1.866
	.2800	-.743	-1.053	-1.286	-1.460	-.811	-1.158	-1.456	-1.628
	.4300	-.650	-1.050	-1.204	-1.407	-.740	-1.100	-1.321	-1.515
	.5800	-.723	-1.102	-1.208	-1.472	-.698	-1.117	-1.250	-1.473
	.7300	-.800	-1.200	-1.300	-1.650	.800	-1.150	-1.350	-1.570
Lower	.0025	.150	-.232	-.302	-.310	.126	-.400	-.473	-.750
	.0050	.278	-.033	-.204	-.200	.172	-.167	.300	-.448
	.0100	.350	.100	.237	-.089	.210	-.067	.169	-.134
	.0250	.500	.400	.457	.300	.500	.330	.405	.370
	.0400	.550	.460	.486	.450	.550	.500	.473	.540
	.0700	.550	.460	.559	.536	.550	.550	.578	.582
	.1700	.449	.325	.465	.370	.534	.388	.481	.226
	.2800	.408	.378	.351	.250	.492	.400	.384	.234
	.4300	.355	.415	.339	.190	.420	.433	.384	.167
	.5800	.392	.423	.359	.165	.412	.450	.388	.138
	.7200	.604	.480	.367	.093	.609	.496	.388	.059
FLAP or AILERON									
Upper	.7310	.400	.500	.400	-.350	.450	.500	.440	-.500
	.7395	-4.511	-4.923	-4.556	-4.459	-3.857	-4.563	-4.173	-3.397
	.7532	-5.074	-5.589	-5.266	-5.536	-4.286	-5.084	-4.784	-3.883
	.7806	-1.388	-1.622	-1.878	-2.451	-1.105	-1.421	-1.629	-1.653
	.8354	-.576	-.589	-.763	-1.133	-.424	-.504	-.633	-.787
	.9038	-.188	-.419	-.359	-.589	-.101	-.358	-.295	-.515
	.9440	-.004	-.431	-.118	-.375	.042	-.342	-.122	-.427
	.9863	.200	-.264	-.020	-.290	.150	-.200	-.030	-.397
	.9928	.273	-.248	.082	-.190	.277	-.175	.068	-.360
Lower	.7312	.669	.550	.420	.180	.681	.600	.447	.188
	.7367	.669	.573	.420	.173	.681	.579	.450	.163
	.7428	.669	.561	.420	.177	.681	.579	.451	.167
	.7532	.608	.545	.420	.141	.680	.554	.451	.121
	.7806	.678	.512	.445	.177	.710	.525	.481	.146
	.8628	.649	.435	.404	.145	.668	.488	.439	.117
	.9313	.567	.398	.320	-.050	.584	.350	.370	-.030
SLAT									
Upper	-.0200		-1.550	-2.250	-1.838		-2.200	-2.650	-2.330
	-.0700		-2.421	-1.958	-1.050		-3.149	-2.471	-1.700
	-.1250		-2.484	-1.754	-1.008		-3.640	-2.715	-1.731
	-.1400		-3.971	-3.129	-1.304		-5.921	-5.392	-2.405
	-.1450		-5.446	-2.825	-1.258		-8.566	-4.281	-2.450
	-.1500		-5.638	-3.304	-.854		-10.499	-7.487	-3.078

TABLE 14

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 6.2°				α = 9.8°			
WING									
Upper	.0000	.374	.404	.330	.342	.235	-.474	-1.672	-1.409
	.0025	.429	.217	-.046	-.056	-.147	-1.008	-2.331	-1.986
	.0150	.229				-.218			
	.0250	.137	-.072	-.297	-.350	-.231	-.718	-1.193	-1.219
	.0400	.099	-.182	-.322	-.326	-.247	-.768	-1.033	-1.008
	.0700	-.062	-.310	-.351	-.374	-.386	-.735	-.941	-.956
	.1200	-.379	-.565	-.665	-.709	-.693	-.978	-1.188	-1.258
	.1900	-.999	-1.229	-1.234	-1.592	-1.298	-1.617	-1.545	-2.055
	.2200	-.704	-.970	-.987	-1.008	-.924	-1.293	-1.407	-1.383
	.2800	-.641	-.812	-.828	-.810	-.814	-1.004	-1.167	-1.111
	.4300	-.504	-.595	-.560	-.487	-.613	-.697	-.693	-.633
	.5800	-.512	-.527	-.410	-.338	-.571	-.567	-.491	-.443
	.6800	-.574	-.531	-.334	-.274	-.600	-.525	-.394	-.362
Lower	.0025	.087	.085	.276	.185	.436	.386	.159	.112
	.0050	.124	.161	.326	.217	.403	.331	.298	.297
	.0100	.145	.178	.305	.169	.352	.449	.378	.374
	.0250	-.025	.238	.267	.185	.226	.432	.399	.366
	.0700	.254	.191	.309	.201	.352	.373	.378	.288
	.1200	.320	.365	.376	.274	.403	.281	.415	.323
	.2200	.066	.446	.330	.258	.063	.474	.369	.280
	.2800	.320	.285	.276	.201	.369	.323	.323	.245
	.4300	.291	.302	.121	.052	.323	.352	.180	.099
	.5800	.416	.319	.008	-.044	.436	.352	.058	-.012
	.7200	.533	.374	-.025	.004	.541	.373	.058	.034
FLAP or AILERON									
Upper	.7312	-.090	-.076			-.087	-.067		
	.7395	-.099	-.088			-.093	-.074		
	.7532	-.591	-.506	-.021	-.016	-.592	-.441	-.031	-.028
	.7806	-.574	-.502	-.297	-.197	-.567	-.432	-.373	-.306
	.8354	-.587	-.497	-.238	-.137	-.583	-.432	-.331	-.258
	.9038	-.670	-.497	-.188	-.092	-.646	-.432	-.298	-.198
	.9440	-.608	-.514	-.192	-.092	-.588	-.453	-.273	-.181
	.9863	-.550	-.480	-.142	-.060	-.500	-.400	-.250	-.170
	.9928	-.491	-.446	-.108	-.056	-.466	-.394	-.218	-.155
Lower	.7312	.612	.463	-.008	-.104	.621	.453	.130	-.021
	.7367	.595	.451	.012	-.088	.604	.453	.147	-.004
	.7428	.591	.489	.046	.016	.604	.470	.189	.185
	.7532	.591	.459	.389	.366	.600	.441	.424	.452
	.7806	.620	.429	.025	-.040	.621	.411	.163	.047
	.8628	.524	.327	.012	-.040	.525	.310	.100	.017
	.9313	.345	.157	-.025	-.020	.348	.130	.025	-.017

TABLE 14 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 13.4°				α = 17.3°			
WING									
Upper	.0000	-1.175	-3.843	-6.003	-4.853	-3.738	-7.299	-7.194	-4.834
	.0025	-1.679	-4.077	-6.206	-4.671	-4.031	-7.003	-7.022	-4.490
	.0150	-.947				-1.908			
	.0250	-.833	-1.834	-2.355	-2.017	-1.573	-2.454	-2.236	-1.727
	.0400	-.745	-1.678	-1.927	-1.579	-1.330	-2.086	-1.781	-1.290
	.0700	-.798	-1.389	-1.661	-1.375	-1.261	-1.558	-1.486	-1.071
	.1200	-1.135	-1.628	-1.698	-1.420	-1.532	-1.613	-1.149	-.910
	.1900	-1.675	-2.201	-1.720	-1.282	-2.059	-1.604	-.686	-.629
	.2200	-1.223	-2.077	-2.170	-1.251	-1.788	-1.636	-.886	-.620
	.2800	-1.039	-1.467	-2.202	-1.198	-1.330	-1.495	-.886	-.620
	.4300	-.736	-.871	-1.333	-1.070	-.876	-1.199	-.836	-.620
	.5800	-.675	-.665	-.666	-.924	-.899	-1.104	-.799	-.611
	.6800	-.719	-.573	-.499	-.827	-1.087	-1.008	-.804	-.616
Lower	.0025	.135	-.688	-.662	-.911	-.683	-2.036	-1.004	-.986
	.0050	.381	-.018	-.351	-.150	-.110	-.559	-.618	-.147
	.0100	.412	.325	.004	.199	.229	-.036	-.145	.187
	.0250	.359	.499	.364	.367	.380	.440	.331	.366
	.0700	.464	.582	.414	.327	.499	.668	.431	.339
	.1200	.464	.266	.472	.353	.486	.190	.459	.339
	.2200	.070	.555	.405	.305	.068	.563	.395	.299
	.2800	.429	.403	.355	.269	.472	.431	.354	.276
	.4300	.403	.408	.202	.110	.440	.413	.172	.116
	.5800	.482	.403	.081	.000	.509	.381	.013	-.013
	.7200	.583	.394	.004	-.026	.614	.368	-.031	-.075
FLAP or AILERON									
Upper	.7312	-.099	-.057			-.207	-.132		
	.7395	-.108	-.061			-.309	-.145		
	.7532	-.714	-.399	-.036	-.043	-1.756	-1.136	-.089	-.075
	.7806	-.670	-.399	-.409	-.601	-1.729	-1.049	-.881	-.629
	.8354	-.705	-.389	-.346	-.566	-1.513	-.913	-.795	-.616
	.9038	-.732	-.366	-.279	-.539	-.766	-.654	-.672	-.584
	.9440	-.627	-.385	-.229	-.504	-.568	-.604	-.650	-.531
	.9863	-.550	-.360	-.207	-.468	-.500	-.500	-.590	-.535
	.9928	-.499	-.344	-.162	-.371	-.344	-.490	-.581	-.544
Lower	.7312	.662	.454	.031	-.097	.669	.440	.063	-.160
	.7367	.644	.444	.067	-.079	.678	.436	.095	-.147
	.7428	.644	.477	.099	.128	.678	.481	.145	.147
	.7532	.640	.440	.409	.424	.655	.431	.463	.473
	.7806	.653	.394	.076	-.044	.660	.377	.086	-.116
	.8628	.552	.261	.031	-.106	.582	.249	-.013	-.178
	.9313	.385	.100	-.018	-.190	.463	.072	-.159	-.267

TABLE 15

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .178 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.0 °				α = 8.6 °			
WING									
Upper	.0000	.398	-.661	-1.874	-1.034	-.530	-3.709	-5.705	-4.466
	.0025	.097	-1.190	-2.535	-1.389	-1.071	-3.913	-5.994	-4.735
	.0150	-.111				-.683			
	.0250	-.145	-.837	-1.357	-1.113	-.688	-1.857	-2.461	-2.314
	.0400	-.184	-.904	-1.217	-.926	-.663	-1.734	-2.076	-1.878
	.0700	-.378	-.895	-1.139	-.931	-.770	-1.515	-1.827	-1.690
	.1200	-.723	-1.204	-1.449	-1.310	-1.147	-1.821	-2.040	-1.928
	.1900	-1.514	-2.099	-2.023	-2.305	-1.892	-2.683	-2.355	-2.558
	.2200	-1.155	-1.718	-1.743	-1.502	-1.428	-2.249	-2.289	-2.162
	.2800	-1.067	-1.461	-1.482	-1.236	-1.316	-1.852	-1.959	-1.867
	.4300	-1.009	-1.328	-1.043	-.778	-1.178	-1.556	-1.243	-.999
	.5800	-1.315	-1.542	-.787	-.571	-1.438	-1.755	-.908	-.715
	.6800	-1.941	-2.223	-.647	-.482	-2.056	-2.464	-.746	-.609
Lower	.0025	.388	.280	.072	.192	.306	-.607	-.619	-.868
	.0050	.349	.299	.217	.295	.367	-.071	-.360	-.152
	.0100	.291	.461	.270	.295	.387	.280	-.065	.147
	.0250	.155	.457	.313	.246	.290	.438	.274	.274
	.0700	.320	.438	.284	.162	.392	.551	.329	.213
	.1200	.388	.450	.328	.226	.403	.550	.370	.218
	.2200	.390	.504	.251	.192	.400	.520	.284	.203
	.2800	.378	.309	.202	.147	.403	.280	.233	.162
	.4300	.378	.361	-.004	-.044	.403	.362	.015	-.025
	.5800	.509	.309	-.198	-.137	.515	.301	-.162	-.137
	.7200	.606	.114	-.222	-.118	.617	.107	-.208	-.137
FLAP or AILERON									
Upper	.7312	.031	.023			.036	.020		
	.7395	-2.076	-1.456			-2.139	-1.558		
	.7532	-18.508	-23.231	-.119	-.091	-19.793	-25.317	-.113	-.091
	.7806	-3.286	-10.824	-1.729	-.768	-3.377	-11.449	-1.639	-.802
	.8354	-1.650	-3.951	-.782	-.428	-1.688	-4.142	-.781	-.492
	.9038	-.563	-2.028	-.468	-.310	-.596	-2.158	-.507	-.375
	.9440	-.572	-2.094	-.637	-.330	-.653	-2.209	-.690	-.406
	.9863	-.400	-2.151	-.565	-.300	-.700	-2.392	-.578	-.355
	.9928	-.300	-1.447	-.299	-.241	-.346	-1.551	-.324	-.314
Lower	.7312	.626	.195	-.313	-.266	.617	.183	-.289	-.279
	.7367	.650	.171	-.280	-.251	.642	.158	-.243	-.253
	.7428	.674	.223	-.280	-.251	.678	.224	-.253	-.253
	.7532	.684	.171	.280	.275	.673	.153	.324	.294
	.7806	.684	.095	-.270	-.206	.673	.076	-.248	-.238
	.8628	.640	-.185	-.260	-.211	.617	-.219	-.258	-.228
	.9313	.563	-.499	-.260	-.177	.561	-.576	-.258	-.228

TABLE 15 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60^{o}$ $\delta_a = 00^{o}$ $i_t = 00^{o}$

$C_{\mu_f} =$ .178 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.3 °				α = 16.3 °			
WING									
Upper	.0000	-2.553	-8.296	-13.152	-9.848	-6.682	-14.188	-3.469	-3.049
	.0025	-2.999	-7.899	-12.578	-8.186	-6.538	-9.683	-3.420	-2.712
	.0150	-1.531				-2.866			
	.0250	-1.351	-3.015	-4.122	-3.822	-2.310	-5.827	-3.360	-2.646
	.0400	-1.180	-2.634	-3.441	-3.031	-1.916	-4.584	-3.349	-2.585
	.0700	-1.196	-2.164	-2.978	-2.536	-1.777	-2.911	-3.349	-2.552
	.1200	-1.553	-2.444	-2.696	-2.301	-2.049	-2.574	-3.196	-2.231
	.1900	-2.265	-3.164	-2.494	-2.281	-2.655	-3.154	-1.830	-1.568
	.2200	-1.744	-2.867	-2.989	-2.150	-2.399	-2.883	-2.207	-1.530
	.2800	-1.531	-2.206	-2.733	-1.999	-1.922	-2.529	-2.081	-1.519
	.4300	-1.281	-1.777	-1.691	-1.562	-1.383	-2.043	-1.863	-1.469
	.5800	-1.526	-1.947	-1.170	-1.249	-1.566	-2.043	-1.726	-1.403
	.6800	-2.127	-2.661	-.941	-1.067	-2.133	-2.275	-1.600	-1.408
Lower	.0025	-.234	-2.470	-2.228	-2.598	-1.844	-4.927	-1.136	-1.469
	.0050	.111	-.772	-1.728	-.994	-.716	-1.745	-.950	-.552
	.0100	.287	-.243	-.872	-.286	-.016	-.828	-.475	-.127
	.0250	.340	.338	.106	.171	.322	.165	.180	.171
	.0700	.452	.640	.329	.244	.438	.668	.360	.226
	.1200	.452	.550	.388	.249	.516	.600	.409	.226
	.2200	.450	.513	.313	.203	.530	.508	.327	.182
	.2800	.446	.333	.249	.171	.538	.508	.300	.149
	.4300	.452	.375	.053	.010	.538	.403	.092	-.022
	.5800	.542	.291	-.127	-.140	.599	.325	-.092	-.165
	.7200	.664	.121	-.212	-.203	.716	.198	-.234	-.259
FLAP or AILERON									
Upper	.7312	.028	.019			.036	.036		
	.7395	-2.194	-1.589			-2.186	-.931		
	.7532	-20.090	-26.476	-.118	-.107	-19.104	-19.405	-.156	-.151
	.7806	-3.292	-12.158	-1.563	-1.130	-3.071	-7.076	-1.797	-1.408
	.8354	-1.606	-4.317	-.909	-.880	-1.516	-2.082	-1.398	-1.314
	.9038	-.537	-2.243	-.654	-.765	-.422	-1.497	-1.251	-1.220
	.9440	-.558	-2.338	-.659	-.718	-.461	-1.248	-.650	-1.088
	.9863	-.500	-2.433	-.659	-.703	-.450	-1.176	-1.070	-1.126
	.9928	-.281	-1.677	-.468	-.578	-.172	-1.126	-1.021	-1.126
Lower	.7312	.638	.195	-.308	-.281	.655	.270	-.300	-.359
	.7367	.680	.185	-.265	-.260	.699	.248	-.278	-.353
	.7428	.718	.253	-.265	-.265	.744	.320	-.278	-.336
	.7532	.718	.179	.329	.369	.738	.265	.344	.331
	.7806	.718	.079	-.281	-.255	.727	.182	-.295	-.331
	.8628	.654	-.238	-.303	-.317	.672	-.027	-.398	-.452
	.9313	.595	-.587	-.340	-.369	.633	-.265	-.519	-.574

TABLE 16

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .180 $C_{\mu,LE}$: I = .004 ,C = .007 ,O = .023 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.0°				α = 8.6°			
WING									
Upper	.0000	.401	-.712	-2.445	-1.786	-.786	-3.829	-6.766	-5.464
	.0025	.616	-1.397	-3.298	-2.881	-.756	-4.108	-7.192	-6.775
	.0150	.060				-.716			
	.0250	-.079	-1.018	-1.303	-.922	-.716	-1.938	-2.415	-2.280
	.0400	-.107	-.833	-1.156	.299	-.626	-1.611	-2.004	-.382
	.0700	-.397	-.823	-.853	-.581	-.820	-1.345	-1.485	-1.489
	.1200	-.714	-1.198	-1.431	-1.372	-1.184	-1.715	-2.009	-2.224
	.1900	-1.424	-2.115	-2.208	-2.627	-1.910	-2.563	-2.648	-3.658
	.2200	-1.144	-1.661	-1.786	-1.608	-1.417	-2.004	-2.118	-2.214
	.2800	-1.083	-1.499	-1.464	-1.236	-1.328	-1.739	-1.717	-1.704
	.4300	-.999	-1.333	-1.066	-.781	-1.189	-1.440	-1.232	-1.045
	.5800	-1.280	-1.564	-.781	-.563	-1.442	-1.658	-.895	-.801
	.6800	-1.924	-2.282	-.720	-.563	-2.074	-2.341	-.796	-.795
Lower	.0025	.355	.277	.014	.081	.248	-.715	-.831	-1.260
	.0050	.317	.291	.151	.259	.323	-.028	-.529	-.270
	.0100	.270	.411	.236	.290	.398	.241	-.178	.096
	.0250	.135	.425	.303	.268	.288	.478	.227	.255
	.0700	.308	.398	.293	.186	.393	.563	.297	.219
	.1200	.373	.189	.322	.231	.412	.550	.336	.234
	.2200	.360	.490	.246	.209	.400	.511	.257	.198
	.2800	.359	.400	.208	.149	.393	.383	.212	.147
	.4300	.369	.347	-.004	-.013	.398	.360	.014	-.020
	.5800	.504	.277	-.208	-.127	.512	.293	-.168	-.132
	.7200	.598	.092	-.241	-.095	.616	.142	-.227	-.137
FLAP or AILERON									
Upper	.7312	.071	-.086			-.095	-.094		
	.7395	-2.066	-1.476			-2.149	-1.550		
	.7532	-18.633	-24.372	-.062	-.040	-19.110	-25.855	-.092	-.074
	.7806	-3.326	-11.387	-1.483	-.886	-3.417	-11.380	-1.767	-.882
	.8354	-1.653	-4.369	-.710	-.445	-1.681	-4.212	-.821	-.520
	.9038	-.579	-2.272	-.431	-.281	-.601	-2.213	-.504	-.392
	.9440	-.635	-2.495	-.687	-.413	-.681	-2.336	-.747	-.408
	.9863	-.600	-2.629	-.635	-.349	-.600	-2.402	-.658	-.382
	.9928	-.336	-1.809	-.331	-.218	-.393	-1.563	-.321	-.316
Lower	.7312	.635	.083	-.345	-.240	.621	.165	-.287	-.260
	.7367	.654	.060	-.322	-.231	.636	.161	-.262	-.239
	.7428	.677	-.004	-.312	-.259	.621	.213	-.247	-.265
	.7532	.691	.037	-.336	-.213	.686	.142	-.272	-.244
	.7806	.696	.013	-.312	-.199	.686	.075	-.252	-.188
	.8628	.644	-.356	-.293	-.199	.611	-.232	-.257	-.204
	.9313	.588	-.657	-.270	-.154	.582	-.578	-.247	-.193

TABLE 16 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .180 $C_{\mu,LE}$: I = .004 ,C = .007 ,O = .023 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of - 0.31, α = 12.2°	0.56, α = 12.2°	0.76, α = 12.2°	0.90, α = 12.2°	0.31, α = 15.8°	0.56, α = 15.8°	0.76, α = 15.8°	0.90, α = 15.8°
WING									
Upper	.0000	-2.659	-8.243	-14.205	-11.426	-6.152	-15.814	-25.444	-22.523
	.0025	-2.577	-8.037	-13.753	-12.557	-5.457	-12.402	-24.068	-23.142
	.0150	-1.484				-2.547			
	.0250	-1.324	-3.038	-3.980	-3.766	-2.131	-4.899	-6.323	-5.765
	.0400	-1.097	-2.479	-3.184	-1.370	-1.721	-3.933	-4.886	-2.845
	.0700	-1.185	-1.935	-2.358	-2.345	-1.657	-2.938	-3.664	-3.385
	.1200	-1.571	-2.308	-2.721	-2.913	-1.999	-3.308	-3.815	-3.705
	.1900	-2.247	-3.048	-3.074	-4.248	-1.950	-3.894	-3.778	-4.940
	.2200	-1.613	-2.357	-2.532	-2.558	-1.836	-3.027	-3.140	-2.935
	.2800	-1.499	-1.989	-2.049	-1.944	-1.642	-2.518	-2.583	-2.255
	.4300	-1.247	-1.602	-1.417	-1.187	-1.342	-1.977	-1.686	-1.385
	.5800	-1.489	-1.808	-1.024	-.918	-1.578	-2.099	-1.210	-1.045
	.6800	-2.092	-2.504	-.905	-.873	-2.084	-2.894	-1.026	-.970
Lower	.0025	-.273	-2.411	-2.487	-3.725	-1.594	-5.927	-5.399	-7.570
	.0050	.072	-.744	-1.935	-1.385	-.668	-2.204	-4.356	-3.205
	.0100	.309	-.254	-1.034	-.451	-.005	-1.242	-2.551	-1.545
	.0250	.324	.367	.029	.131	.268	.027	-.394	-.120
	.0700	.463	.642	.268	.213	.421	.651	.248	.240
	.1200	.474	.600	.333	.223	.463	.600	.362	.270
	.2200	.450	.519	.258	.182	.470	.502	.308	.245
	.2800	.438	.392	.213	.152	.478	.458	.259	.210
	.4300	.458	.387	.034	.020	.484	.364	.070	.075
	.5800	.561	.318	-.124	-.121	.542	.287	-.097	-.040
	.7200	.649	.147	-.199	-.111	.631	.121	-.151	-.070
FLAP or AILERON									
Upper	.7312	-.094	-.096			-.094	-.109		
	.7395	-2.175	-1.628			-2.177	-1.791		
	.7532	-18.657	-26.019	-.099	-.075	-19.100	-27.391	-.181	-.122
	.7806	-3.350	-11.492	-1.781	-.832	-3.289	-11.841	-2.513	-1.195
	.8354	-1.613	-4.376	-.885	-.568	-1.568	-4.750	-1.140	-.715
	.9038	-.597	-2.269	-.557	-.446	-.563	-2.563	-.670	-.535
	.9440	-.680	-2.401	-.796	-.426	-.689	-2.684	-.799	-.535
	.9863	-.650	-2.514	-.701	-.406	-.600	-2.844	-.843	-.485
	.9928	-.324	-1.632	-.358	-.360	-.342	-1.922	-.389	-.350
Lower	.7312	.659	.186	-.288	-.218	.647	.204	-.189	-.145
	.7367	.690	.176	-.263	-.208	.657	.176	-.178	-.140
	.7428	.711	.230	-.258	-.228	.684	.254	-.156	-.105
	.7532	.711	.176	-.273	-.213	.684	.171	-.178	-.130
	.7806	.700	.102	-.253	-.177	.678	.077	-.156	-.105
	.8628	.649	-.249	-.268	-.203	.631	-.281	-.205	-.160
	.9313	.592	-.607	-.268	-.203	.573	-.635	-.216	-.190

TABLE 17

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60 ° δ_a = 00 ° i_t = 00 °

C_{μ_f} = .187 $C_{\mu,LE}$: I = .000 , C = .000 , O = .000 $C_{\mu,K}$: I = .003 , C = .007 , O = .021

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 4.9 °				α = 8.6 °			
WING									
Upper	.0000	.400	-.834	-2.317	-1.551	-.492	-3.741	-6.265	-5.269
	.0025	.400	-1.341	-2.974	-2.395	-1.036	-3.984	-6.479	-5.916
	.0150	-.100				-.714			
	.0250	-.150	-.917	-1.595	-1.273	-.663	-1.899	-2.607	-2.523
	.0400	-.200	-.980	-1.378	-1.082	-.642	-1.799	-2.193	-2.062
	.0700	-.350	-.951	-1.282	-1.053	-.766	-1.578	-1.964	-1.880
	.1200	-.800	-1.273	-1.641	-1.453	-1.170	-1.905	-2.219	-2.201
	.1900	-1.400	-1.965	-1.919	-2.414	-1.802	-2.499	-2.362	-3.243
	.2200	-1.050	-1.697	-1.772	-1.629	-1.372	-2.163	-2.086	-2.119
	.2800	-1.050	-1.463	-1.474	-1.156	-1.352	-1.805	-1.704	-1.543
	.4300	-1.000	-1.321	-1.090	-.726	-1.170	-1.526	-1.229	-.974
	.5800	-1.300	-1.629	-.737	-.595	-1.450	-1.873	-.826	-.777
	.6800	-1.950	-2.302	-.691	-.604	-2.051	-2.547	-.775	-.756
Lower	.0025	.400	.273	.020	.121	.336	-.699	-.739	-.917
	.0050	.400	.321	.156	.248	.373	-.047	-.479	-.165
	.0100	.350	.458	.242	.282	.398	.284	-.127	.129
	.0250	.200	.497	.308	.239	.295	.478	.234	.279
	.0700	.340	.434	.303	.190	.414	.563	.311	.207
	.1200	.400	.450	.328	.234	.419	.550	.331	.238
	.2200	.400	.507	.257	.180	.400	.526	.275	.196
	.2800	.400	.317	.191	.141	.404	.299	.224	.160
	.4300	.550	.356	-.030	-.048	.404	.363	.025	-.015
	.5800	.550	.307	-.217	-.141	.518	.299	-.158	-.129
	.7200	.650	.102	-.247	-.112	.616	.115	-.198	-.129
FLAP or AILERON									
Upper	.7312	-.400	-.099			-.098	-.112		
	.7395	-22.100	-1.614			-2.267	-1.769		
	.7532	-19.750	-24.538	-.069	-.039	-20.619	-26.978	-.111	-.079
	.7806	-3.400	-10.426	-2.004	-1.087	-3.424	-11.137	-2.367	-1.429
	.8354	-1.650	-4.092	-.919	-.497	-1.652	-4.284	-1.071	-.673
	.9038	-.600	-2.034	-.499	-.312	-.606	-2.173	-.535	-.398
	.9440	-.650	-2.034	-.717	-.453	-.657	-2.115	-.729	-.533
	.9863	-.400	-2.068	-.742	-.385	-.650	-2.157	-.775	-.492
	.9928	-.350	-1.380	-.363	-.278	-.341	-1.473	-.484	-.347
Lower	.7312	.650	.204	-.313	-.268	.652	.199	-.275	-.274
	.7367	.700	.185	-.292	-.258	.673	.184	-.234	-.248
	.7428	.700	.234	-.292	-.239	.709	.268	-.198	-.233
	.7532	.750	.151	-.308	-.229	.694	.152	-.239	-.259
	.7806	.700	.102	-.277	-.204	.689	.121	-.209	-.212
	.8628	.700	-.185	-.267	-.185	.652	-.210	-.209	-.212
	.9313	.500	-.507	-.267	-.175	.595	-.552	-.234	-.217

TABLE 17 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .187 $C_{\mu,LE}$: I = .000 , C = .000 , O = .000 $C_{\mu,K}$: I = .003 , C = .007 , O = .021

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.2°				α = 15.9°			
WING									
Upper	.0000	-2.983	-8.640	-13.767	12.106	-6.138	-15.495	-6.265	-6.866
	.0025	-3.357	-8.179	-13.214	12.214	-6.049	-10.808	-6.232	-6.595
	.0150	-1.705				-2.738			
	.0250	-1.454	-3.047	-4.411	-4.207	-2.222	-6.186	-6.287	-6.916
	.0400	-1.261	-2.687	-3.673	-3.350	-1.877	-4.919	-6.330	-6.866
	.0700	-1.261	-2.206	-3.192	-2.936	-1.766	-3.238	-6.422	-6.750
	.1200	-1.684	-2.470	-3.111	-3.010	-2.116	-2.971	-6.107	-5.198
	.1900	-2.138	-2.931	-2.935	-3.941	-2.483	-3.527	-2.657	-4.027
	.2200	-1.593	-2.455	-2.587	-2.521	-1.844	-2.948	-2.689	-2.762
	.2800	-1.513	-2.031	-2.096	-1.829	-1.716	-2.437	-2.260	-2.121
	.4300	-1.267	-1.661	-1.438	-1.154	-1.394	-1.942	-1.586	-1.408
	.5800	-1.502	-1.952	-.989	-.893	-1.599	-2.170	-1.152	-1.110
	.6800	-2.117	-2.566	-.903	-.893	-2.172	-2.829	-1.005	-1.082
Lower	.0025	-.374	-2.656	-2.464	-3.265	-1.638	-5.675	-2.401	-4.032
	.0050	.021	-.814	-1.924	-1.351	-.661	-2.062	-2.108	-2.005
	.0100	.262	-.259	-1.026	-.489	.005	-1.102	-1.298	-1.010
	.0250	.336	.370	.026	.143	.288	.119	-.141	-.055
	.0700	.475	.677	.315	.234	.433	.670	.255	.187
	.1200	.454	.600	.363	.244	.477	.550	.331	.254
	.2200	.450	.550	.304	.202	.490	.488	.293	.204
	.2800	.470	.407	.240	.148	.499	.357	.255	.171
	.4300	.465	.417	.042	-.005	.499	.374	.059	-.005
	.5800	.577	.317	-.128	-.132	.588	.295	-.092	-.143
	.7200	.663	.179	-.192	-.148	.677	.147	-.195	-.154
FLAP or AILERON									
Upper	.7312	-.107	-.106			-.097	-.122		
	.7395	-2.305	-1.722			-2.376	-1.836		
	.7532	-20.525	-27.519	-.224	-.162	-21.008	-27.756	-.157	-.131
	.7806	-3.406	-11.682	-3.138	-1.808	-3.449	-11.328	-2.380	-1.740
	.8354	-1.577	-4.417	-1.331	-.819	-1.605	-4.374	-1.086	-.933
	.9038	-.582	-2.164	-.684	-.510	-.555	-2.153	-.690	-.651
	.9440	-.620	-2.132	-.855	-.611	-.666	-2.261	-.847	-.734
	.9863	-.600	-2.285	-.951	-.601	-.600	-2.363	-.820	-.684
	.9928	-.310	-1.507	-.433	-.494	-.349	-1.641	-.516	-.486
Lower	.7312	.668	.232	-.224	-.234	.644	.227	-.239	-.254
	.7367	.700	.195	-.203	-.202	.688	.221	-.217	-.237
	.7428	.727	.301	-.176	-.202	.722	.312	-.211	-.259
	.7532	.716	.206	-.197	-.218	.705	.227	-.239	-.243
	.7806	.732	.111	-.181	-.186	.722	.119	-.233	-.220
	.8628	.673	-.248	-.219	-.202	.655	-.193	-.244	-.276
	.9313	.620	-.566	-.256	-.228	.616	-.494	-.271	-.265

TABLE 18

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

δ_f = 60 ° δ_a = 00 ° i_t = 00 °

C_{μ_f} = .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 6.2 °				α = 9.8 °			
WING									
Upper	.0000	.234	.370	.402	.370	.492	.009	-.657	-.460
	.0025	.475	.346	.262	.280	.273	-.523	-1.251	-1.000
	.0150	.348				.041			
	.0250	.262	.049	-.098	-.086	-.017	-.498	-.774	-.799
	.0400	.197	-.081	-.156	-.123	-.054	-.574	-.703	-.644
	.0700	.016	-.215	-.205	-.202	-.219	-.613	-.665	-.644
	.1200	-.295	-.480	-.557	-.580	-.578	-.889	-.975	-.987
	.1900	-1.045	-1.297	-1.061	-1.539	-1.289	-1.685	-1.276	-1.674
	.2200	-.729	-.992	-.975	-1.033	-.905	-1.349	-1.544	-1.552
	.2800	-.656	-.801	-.807	-.815	-.793	-1.013	-1.243	-1.397
	.4300	-.508	-.589	-.545	-.494	-.578	-.694	-.628	-.657
	.5800	-.512	-.488	-.389	-.346	-.562	-.566	-.439	-.418
	.6800	-.582	-.488	-.316	-.276	-.587	-.532	-.347	-.318
Lower	.0025	-.086	.122	.213	-.004	.397	.391	.289	.285
	.0050	-.025	.138	.246	.058	.351	.353	.410	.385
	.0100	.102	.154	.258	.058	.314	.438	.435	.385
	.0250	-.033	.285	.258	.140	.190	.438	.410	.343
	.0700	.266	.228	.311	.214	.368	.391	.389	.293
	.1200	.320	.378	.389	.296	.434	.340	.427	.343
	.2200	.340	.472	.352	.276	.430	.511	.377	.301
	.2800	.357	.358	.307	.230	.417	.374	.343	.259
	.4300	.299	.341	.168	.074	.368	.374	.201	.130
	.5800	.438	.341	.041	-.016	.483	.374	.075	.017
	.7200	.557	.382	.033	.016	.595	.400	.088	.059
FLAP or AILERON									
Upper	.7312	-.090	-.062			-.085	-.070		
	.7395	-.100	-.071			-.090	-.075		
	.7532	-.586	-.382	-.019	-.018	-.541	-.443	-.025	-.021
	.7806	-.557	-.354	-.275	-.243	-.512	-.421	-.318	-.264
	.8354	-.561	-.354	-.229	-.189	-.537	-.421	-.285	-.201
	.9038	-.635	-.354	-.189	-.123	-.587	-.421	-.255	-.163
	.9440	-.578	-.374	-.184	-.119	-.529	-.434	-.222	-.130
	.9863	-.550	-.240	-.127	-.095	-.480	-.400	-.121	-.096
	.9928	-.467	-.341	-.107	-.095	-.413	-.379	-.167	-.105
Lower	.7312	.631	.435	.070	-.041	.661	.485	.176	-.004
	.7367	.631	.435	.082	-.021	.661	.485	.184	-.004
	.7428	.627	.463	.127	.086	.657	.519	.234	.243
	.7532	.627	.427	.094	-.016	.653	.494	.201	.004
	.7806	.627	.386	.094	.016	.653	.455	.201	.050
	.8628	.537	.297	.074	.016	.570	.362	.159	.021
	.9313	.361	.138	.020	-.004	.409	.191	.079	.000

TABLE 18 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .000 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 13.4°				α = 17.4°			
WING									
Upper	.0000	-.394	-2.091	-3.828	-2.251	-2.326	-4.577	-4.469	-2.768
	.0025	-.926	-2.455	-4.119	-2.281	-2.708	-4.638	-4.577	-2.781
	.0150	-.563				-1.347			
	.0250	-.563	-1.273	-1.652	-1.173	-1.129	-1.771	-1.613	-1.271
	.0400	-.532	-1.234	-1.339	-.892	-.996	-1.530	-1.269	-.943
	.0700	-.645	-1.082	-1.133	-.805	-1.000	-1.155	-1.048	-.816
	.1200	-1.004	-1.346	-1.176	-.900	-1.330	-1.220	-.909	-.782
	.1900	-1.688	-2.126	-.991	-.784	-2.017	-1.297	-.530	-.594
	.2200	-1.242	-2.182	-1.317	-.823	-1.832	-1.323	-.769	-.598
	.2800	-1.026	-1.641	-1.274	-.818	-1.244	-1.414	-.774	-.598
	.4300	-.727	-.823	-1.150	-.736	-.781	-1.013	-.756	-.568
	.5800	-.706	-.654	-1.021	-.684	-.807	-.987	-.756	-.572
	.6800	-.788	-.567	-.901	-.658	-1.004	-.922	-.735	-.568
Lower	.0025	.394	-.104	-.236	-.087	-.206	-1.013	-.391	-.293
	.0050	.450	.212	.004	.251	.172	-.134	-.135	.170
	.0100	.429	.446	.219	.368	.348	.263	.148	.345
	.0250	.346	.515	.433	.394	.412	.513	.409	.410
	.0700	.476	.545	.429	.329	.528	.647	.443	.362
	.1200	.476	.550	.481	.351	.519	.620	.487	.384
	.2200	.470	.563	.403	.320	.520	.569	.426	.349
	.2800	.459	.450	.369	.290	.515	.509	.383	.310
	.4300	.424	.407	.223	.121	.481	.427	.213	.153
	.5800	.498	.411	.060	.009	.532	.397	.035	.022
	.7200	.606	.407	.004	-.022	.639	.371	-.030	-.004
FLAP or AILERON									
Upper	.7312	-.120	-.073			-.197	-.124		
	.7395	-.146	-.079			-.291	-.114		
	.7532	-.879	-.463	-.073	-.069	-1.549	-.991	-.075	-.067
	.7806	-.823	-.433	-.609	-.580	-1.519	-.879	-.743	-.559
	.8354	-.827	-.437	-.584	-.576	-1.292	-.771	-.691	-.559
	.9038	-.814	-.411	-.549	-.571	-.717	-.647	-.622	-.537
	.9440	-.667	-.424	-.450	-.489	-.562	-.590	-.600	-.485
	.9863	-.600	-.400	-.421	-.498	-.459	-.560	-.574	-.506
	.9928	-.541	-.372	-.463	-.494	-.339	-.517	-.574	-.506
Lower	.7312	.675	.463	.064	-.100	.695	.435	.017	-.118
	.7367	.675	.463	.077	-.100	.695	.435	.035	-.122
	.7428	.675	.502	.116	.160	.695	.483	.083	.087
	.7532	.658	.463	.082	-.108	.678	.431	.043	-.144
	.7806	.671	.411	.082	-.052	.678	.388	.043	-.070
	.8628	.593	.294	.026	-.134	.605	.263	-.043	-.153
	.9313	.433	.143	-.094	-.212	.463	.065	-.191	-.223

TABLE 19

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .031 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.5 °				α = 9.2 °			
WING									
Upper	.0000	.388	.345	-.306	.000	.264	-1.342	-3.186	-2.307
	.0025	.474	-.093	-.886	-.408	-.162	-1.815	-3.714	-2.642
	.0150	.276				-.231			
	.0250	.185	-.279	-.753	-.610	-.287	-1.120	-1.733	-1.523
	.0400	.129	-.398	-.703	-.547	-.315	-1.157	-1.490	-1.266
	.0700	-.056	-.513	-.703	-.596	-.486	-1.088	-1.336	-1.188
	.1200	-.414	-.832	-1.091	-.995	-.884	-1.430	-1.640	-1.523
	.1900	-1.233	-1.840	-1.671	-2.134	-1.722	-2.393	-1.962	-2.303
	.2200	-.909	-1.469	-1.571	-1.368	-1.264	-2.009	-2.322	-2.087
	.2800	-.832	-1.252	-1.310	-1.112	-1.148	-1.592	-1.916	-1.761
	.4300	-.741	-1.066	-.895	-.682	-.949	-1.291	-1.028	-.853
	.5800	-.935	-1.199	-.667	-.502	-1.116	-1.366	-.748	-.610
	.6800	-1.358	-1.672	-.543	-.408	-1.528	-1.833	-.598	-.486
Lower	.0025	.086	.376	.279	.247	.435	.106	-.135	-.151
	.0050	.134	.288	.384	.251	.389	.236	.065	.170
	.0100	.159	.336	.384	.229	.333	.435	.220	.321
	.0250	.060	.394	.310	.206	.213	.481	.378	.312
	.0700	.315	.323	.306	.179	.389	.481	.374	.266
	.1200	.358	.336	.356	.260	.417	.400	.416	.284
	.2200	.370	.300	.297	.220	.410	.350	.341	.243
	.2800	.384	.288	.247	.170	.403	.296	.294	.216
	.4300	.371	.367	.068	.018	.393	.375	.107	.055
	.5800	.513	.345	-.078	-.090	.505	.347	-.019	-.046
	.7200	.612	.226	-.105	-.054	.602	.204	-.047	-.046
FLAP or AILERON									
Upper	.7312	-.293	-.314			-.310	-.327		
	.7395	-.712	-.756			-.738	-.771		
	.7532	-6.668	-8.618	-.107	-.077	-6.869	-8.679	-.184	-.122
	.7806	-1.845	-3.291	-1.068	-.547	-1.926	-3.333	-1.406	-.638
	.8354	-.715	-1.544	-.557	-.323	-.759	-1.555	-.696	-.404
	.9038	-.224	-1.287	-.411	-.242	-.259	-1.282	-.500	-.307
	.9440	.009	-.849	-.534	-.251	.009	-.838	-.533	-.303
	.9863	-.086	-.726	-.283	-.197	-.218	-.736	-.332	-.248
	.9928	.315	-.442	-.164	-.161	.310	-.407	-.164	-.211
Lower	.7312	.698	.323	-.119	-.157	.685	.329	-.009	-.133
	.7367	.690	.301	-.105	-.157	.685	.287	.009	-.133
	.7428	.685	.354	-.055	-.072	.685	.366	.079	.028
	.7532	.690	.292	-.087	-.157	.685	.306	.028	-.147
	.7806	.685	.230	-.087	-.094	.685	.241	.028	-.083
	.8628	.647	.044	-.087	-.094	.630	.051	-.014	-.110
	.9313	.565	-.159	-.114	-.099	.560	-.143	-.070	-.128

TABLE 19 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60°$ $\delta_a = 00°$ $i_t = 00°$

$C_{\mu_f} =$ .031 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.9°				α = 17.2°			
WING									
Upper	.0000	-1.113	-4.487	-6.407	-4.265	-3.153	-5.803	-4.986	-3.417
	.0025	-1.646	-4.581	-6.421	-4.161	-3.446	-5.748	-5.058	-3.151
	.0150	-.962				-1.707			
	.0250	-.882	-1.986	-2.333	-1.934	-1.437	-2.142	-1.776	-1.413
	.0400	-.811	-1.817	-1.887	-1.502	-1.228	-1.853	-1.390	-1.073
	.0700	-.910	-1.558	-1.591	-1.346	-1.237	-1.422	-1.143	-.927
	.1200	-1.316	-1.835	-1.540	-1.403	-1.572	-1.505	-.982	-.890
	.1900	-2.094	-2.718	-1.188	-1.355	-2.335	-1.546	-.592	-.702
	.2200	-1.599	-2.633	-1.671	-1.398	-2.191	-1.601	-.892	-.702
	.2800	-1.306	-1.925	-1.671	-1.407	-1.563	-1.628	-.892	-.702
	.4300	-1.038	-1.375	-1.568	-1.208	-.958	-1.060	-.865	-.702
	.5800	-1.146	-1.394	-1.380	-1.133	-1.033	-1.083	-.852	-.702
	.6800	-1.500	-1.652	-1.192	-1.062	-1.349	-1.087	-.830	-.720
Lower	.0025	.170	-.958	-.821	-.791	-.488	-1.449	-.511	-.463
	.0050	.330	-.169	-.498	-.100	.000	-.326	-.251	.060
	.0100	.368	.225	-.099	.204	.256	.133	.058	.280
	.0250	.311	.479	.329	.327	.353	.468	.368	.367
	.0700	.476	.620	.404	.289	.502	.656	.421	.321
	.1200	.476	.560	.451	.294	.488	.600	.466	.321
	.2200	.470	.535	.376	.265	.490	.560	.381	.280
	.2800	.467	.347	.333	.232	.498	.514	.345	.248
	.4300	.434	.408	.155	.066	.460	.394	.170	.096
	.5800	.533	.394	-.005	-.085	.521	.353	.009	-.055
	.7200	.623	.305	-.084	-.137	.628	.307	-.090	-.092
FLAP or AILERON									
Upper	.7312	-.296	-.270			-.261	-.112		
	.7395	-.696	-.619			-.650	-.242		
	.7532	-6.395	-6.243	-.182	-.147	-5.865	-2.766	-.091	-.089
	.7806	-1.731	-1.995	-1.277	-1.005	-1.465	-1.294	-.856	-.716
	.8354	-.674	-.943	-.915	-.943	-.572	-.853	-.767	-.688
	.9038	-.198	-.760	-.807	-.891	-.335	-.794	-.722	-.688
	.9440	.047	-.549	-.750	-.806	-.298	-.729	-.700	-.624
	.9863	-.278	-.681	-.671	-.806	-.777	-.839	-.708	-.661
	.9928	.321	-.418	-.577	-.739	-.074	-.647	-.686	-.661
Lower	.7312	.703	.422	-.052	-.242	.688	.385	-.157	-.202
	.7367	.684	.399	-.019	-.246	.684	.376	-.126	-.202
	.7428	.684	.441	.033	.005	.684	.427	-.090	-.092
	.7532	.684	.399	-.014	-.237	.674	.372	-.157	-.216
	.7806	.684	.366	-.014	-.194	.674	.317	-.126	-.151
	.8628	.627	.216	-.075	-.284	.591	.188	-.206	-.248
	.9313	.547	.028	-.197	-.379	.474	-.046	-.336	-.321

TABLE 20

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60°$ $\delta_a = 00°$ $i_t = 00°$

C_{μ_f} = .033 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .001 ,C = .003 ,O = .010

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.5 °				α = 9.1 °			
WING									
Upper	.0000	.330	.336	-.210	-.030	.324	-1.172	-3.141	-2.577
	.0025	.448	-.091	-.768	-.412	-.039	-1.608	-3.625	-2.647
	.0150	.252				-.149			
	.0250	.165	-.276	-.648	-.566	-.206	-.987	-1.665	-1.511
	.0400	.091	-.397	-.618	-.502	-.237	-1.022	-1.436	-1.251
	.0700	-.087	-.509	-.618	-.558	-.399	-.960	-1.299	-1.194
	.1200	-.452	-.819	-1.004	-.957	-.772	-1.273	-1.621	-1.568
	.1900	-1.239	-1.681	-1.510	-2.158	-1.513	-2.044	-2.066	-2.749
	.2200	-.913	-1.362	-1.360	-1.266	-1.087	-1.647	-1.762	-1.608
	.2800	-.874	-1.207	-1.116	-.970	-1.017	-1.379	-1.432	-1.255
	.4300	-.774	-1.009	-.815	-.596	-.846	-1.110	-.978	-.758
	.5800	-.969	-1.155	-.554	-.425	-1.009	-1.216	-.674	-.551
	.6800	-1.391	-1.659	-.481	-.408	-1.394	-1.696	-.581	-.515
Lower	.0025	.074	.379	.326	.287	.487	.154	-.123	-.141
	.0050	.104	.319	.403	.305	.439	.304	.053	.194
	.0100	.148	.362	.390	.275	.364	.454	.225	.308
	.0250	.048	.397	.343	.253	.267	.502	.366	.357
	.0700	.291	.336	.339	.223	.408	.511	.370	.286
	.1200	.343	.362	.395	.296	.447	.520	.414	.313
	.2200	.360	.340	.330	.266	.440	.537	.344	.273
	.2800	.378	.310	.283	.223	.434	.374	.291	.242
	.4300	.361	.379	.090	.064	.425	.405	.115	.097
	.5800	.513	.349	-.026	-.026	.539	.379	-.004	-.009
	.7200	.613	.267	-.043	.004	.623	.264	-.026	.018
FLAP or AILERON									
Upper	.7312	-.287	-.334			-.279	-.341		
	.7395	-.740	-.791			-.721	-.801		
	.7532	-6.786	-8.715	-.065	-.046	-6.578	-8.925	-.106	-.077
	.7806	-1.900	-3.008	-.918	-.571	-1.776	-3.088	-1.110	-.705
	.8354	-.752	-1.504	-.463	-.296	-.680	-1.524	-.568	-.388
	.9038	-.274	-1.164	-.322	-.197	-.215	-1.211	-.396	-.269
	.9440	-.091	-.750	-.459	-.292	-.026	-.793	-.537	-.348
	.9863	-.043	-.634	-.275	-.185	-.202	-.700	-.304	-.229
	.9928	.235	-.362	-.120	-.120	.281	-.388	-.115	-.145
Lower	.7312	.696	.358	-.064	-.107	.706	.366	-.004	-.066
	.7367	.682	.349	-.060	-.094	.697	.348	.009	-.048
	.7428	.678	.401	-.017	-.039	.706	.392	.048	.048
	.7532	.678	.332	-.043	-.086	.693	.339	.009	-.057
	.7806	.704	.280	-.026	-.047	.710	.269	.018	-.009
	.8628	.643	.125	-.026	-.051	.653	.084	.000	-.022
	.9313	.565	-.060	-.051	-.047	.579	-.097	-.022	-.040

TABLE 20 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60^{\circ}$ $\delta_a = 00^{\circ}$ $i_t = 00^{\circ}$

$C_{\mu_f} = .033$ $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .001 ,C = .003 ,O = .010

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.8°				α = 16.5°			
WING									
Upper	.0000	-1.057	-4.466	-8.223	-7.537	-3.672	-10.506	-12.257	-8.331
	.0025	-1.548	-4.577	-8.129	-7.253	-3.883	-8.888	-11.252	-6.071
	.0150	-.863				-1.812			
	.0250	-.787	-1.955	-2.951	-2.960	-1.511	-3.254	-5.786	-5.363
	.0400	-.708	-1.791	-2.433	-2.360	-1.300	-2.781	-4.995	-4.982
	.0700	-.810	-1.542	-2.111	-2.115	-1.256	-2.250	-4.116	-4.309
	.1200	-1.168	-1.835	-2.281	-2.315	-1.596	-2.495	-2.851	-2.888
	.1900	-1.783	-2.506	-2.491	-3.400	-2.063	-2.946	-2.805	-3.628
	.2200	-1.279	-1.964	-2.103	-2.022	-1.462	-2.294	-2.419	-2.229
	.2800	-1.155	-1.635	-1.701	-1.595	-1.300	-1.857	-1.995	-1.780
	.4300	-.920	-1.240	-1.103	-.942	-.982	-1.348	-1.265	-1.063
	.5800	-1.022	-1.311	-.768	-.675	-1.031	-1.353	-.870	-.771
	.6800	-1.380	-1.760	-.652	-.635	-1.341	-1.763	-.679	-.650
Lower	.0025	.195	-.960	-1.205	-1.902	-.726	-3.286	-2.679	-3.103
	.0050	.336	-.142	-.835	-.662	-.103	-1.062	-2.056	-1.287
	.0100	.389	.236	-.304	-.071	.242	-.406	-1.042	-.484
	.0250	.349	.493	.286	.298	.395	.321	.098	.166
	.0700	.491	.635	.397	.324	.543	.714	.400	.309
	.1200	.500	.600	.455	.347	.529	.650	.451	.327
	.2200	.500	.564	.375	.307	.550	.558	.400	.291
	.2800	.495	.450	.339	.280	.561	.480	.381	.265
	.4300	.465	.431	.170	.120	.538	.451	.214	.121
	.5800	.571	.404	.049	.022	.614	.411	.084	.022
	.7200	.655	.284	.027	.009	.691	.330	.028	.004
FLAP or AILERON									
Upper	.7312	-.261	-.339			-.245	-.332		
	.7395	-.684	-.797			-.625	-.769		
	.7532	-6.189	-8.772	-.107	-.082	-5.605	-8.499	-.180	-.123
	.7806	-1.624	-3.009	-1.089	-.804	-1.417	-2.924	-1.298	-.700
	.8354	-.575	-1.458	-.554	-.449	-.489	-1.415	-.679	-.466
	.9038	-.133	-1.151	-.393	-.329	-.081	-1.116	-.502	-.345
	.9440	.040	-.724	-.536	-.404	.130	-.679	-.544	-.323
	.9863	.049	-.595	-.254	-.236	-.238	-.629	-.316	-.242
	.9928	.336	-.320	-.103	-.156	.417	-.272	-.167	-.179
Lower	.7312	.717	.396	.045	-.031	.735	.397	.065	-.018
	.7367	.721	.369	.054	-.022	.735	.393	.079	-.027
	.7428	.726	.431	.125	.080	.740	.446	.135	.148
	.7532	.712	.373	.067	-.018	.731	.379	.070	-.031
	.7806	.717	.307	.076	.004	.722	.317	.088	.004
	.8628	.659	.120	.045	-.027	.682	.116	.023	-.049
	.9313	.606	-.071	.004	-.040	.628	-.067	-.060	-.085

TABLE 21

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60°$ $\delta_a = 00°$ $i_t = 00°$

$C_{\mu_f} =$ .031 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .009 ,O = .018

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of - 0.31 (α = 5.5°)	0.56 (α = 5.5°)	0.76 (α = 5.5°)	0.90 (α = 5.5°)	0.31 (α = 9.1°)	0.56 (α = 9.1°)	0.76 (α = 9.1°)	0.90 (α = 9.1°)
WING									
Upper	.0000	.341	.338	-.392	.119	.271	-1.203	-3.481	-2.688
	.0025	.445	-.091	-1.027	-.379	-.156	-1.694	-3.972	-3.009
	.0150	.258				-.225			
	.0250	.175	-.301	-.793	-.584	-.275	-1.063	-1.856	-1.743
	.0400	.122	-.420	-.752	-.543	-.280	-1.103	-1.597	-1.445
	.0700	-.087	-.548	-.752	-.603	-.463	-1.054	-1.458	-1.381
	.1200	-.406	-.890	-1.176	-1.046	-.853	-1.401	-1.787	-1.784
	.1900	-1.257	-1.625	-1.662	-2.370	-1.665	-2.090	-2.231	-3.110
	.2200	-.961	-1.470	-1.500	-1.310	-1.243	-1.788	-1.884	-1.761
	.2800	-.873	-1.251	-1.239	-1.018	-1.105	-1.504	-1.551	-1.358
	.4300	-.755	-1.073	-.919	-.639	-.894	-1.221	-1.083	-.862
	.5800	-.952	-1.210	-.622	-.443	-1.046	-1.297	-.755	-.647
	.6800	-1.419	-1.744	-.590	-.443	-1.482	-1.815	-.690	-.647
Lower	.0025	.044	.374	.279	.247	.509	.144	-.204	-.266
	.0050	.100	.301	.383	.265	.450	.306	.000	.128
	.0100	.135	.352	.374	.242	.390	.455	.185	.284
	.0250	.013	.397	.324	.219	.312	.491	.343	.307
	.0700	.293	.324	.324	.192	.431	.491	.343	.248
	.1200	.341	.342	.360	.269	.468	.500	.393	.275
	.2200	.360	.300	.302	.237	.460	.536	.333	.234
	.2800	.384	.269	.261	.196	.450	.432	.278	.206
	.4300	.358	.379	.045	.032	.422	.401	.093	.046
	.5800	.511	.356	-.090	-.064	.546	.378	-.037	-.073
	.7200	.620	.233	-.090	-.009	.651	.234	-.060	-.046
FLAP or AILERON									
Upper	.7312	-.317	-.376			-.312	-.371		
	.7395	-.744	-.880			-.741	-.878		
	.7532	-6.981	-9.817	-.072	-.011	-6.825	-9.751	-.086	-.055
	.7806	-1.943	-3.603	-1.149	-.402	-1.794	-3.590	-1.264	-.789
	.8354	-.751	-1.740	-.558	-.219	-.624	-1.712	-.643	-.445
	.9038	-.314	-1.420	-.365	-.155	-.161	-1.410	-.468	-.326
	.9440	-.135	-.991	-.540	-.192	.069	-.950	-.630	-.390
	.9863	-.201	-.886	-.378	-.146	.200	-.513	-.338	-.284
	.9928	.218	-.552	-.198	-.146	.450	-.468	-.190	-.225
Lower	.7312	.685	.301	-.104	-.132	.780	.338	-.060	-.147
	.7367	.668	.301	-.099	-.132	.775	.333	-.042	-.133
	.7428	.664	.333	-.063	-.137	.816	.410	.005	-.101
	.7532	.664	.279	-.086	-.114	.761	.315	-.046	-.128
	.7806	.681	.219	-.063	-.059	.761	.248	-.023	-.078
	.8628	.620	.005	-.077	-.096	.743	.045	-.037	-.110
	.9313	.541	-.201	-.099	-.068	.674	-.167	-.069	-.110

TABLE 21 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60°$ $\delta_a = 00°$ $i_t = 00°$

$C_{\mu_f} = .031$ $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .009 ,O = .018

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.8 °				α = 16.5 °			
WING									
Upper	.0000	-1.127	-4.671	-9.979	-8.013	-3.576	-10.911	-13.262	-8.451
	.0025	-1.676	-4.782	-9.748	-8.219	-3.870	-9.377	-12.137	-6.614
	.0150	-.948				-1.889			
	.0250	-.878	-2.055	-3.462	-3.370	-1.591	-3.531	-6.356	-6.182
	.0400	-.812	-1.889	-2.853	-2.690	-1.360	-3.043	-5.395	-5.845
	.0700	-.906	-1.629	-2.485	-2.408	-1.365	-2.497	-4.390	-5.201
	.1200	-1.305	-1.940	-2.636	-2.643	-1.716	-2.775	-3.072	-3.427
	.1900	-2.075	-2.472	-2.759	-3.901	-2.328	-3.052	-3.028	-4.172
	.2200	-1.591	-2.041	-2.330	-2.244	-2.379	-2.478	-2.594	-2.548
	.2800	-1.296	-1.703	-1.901	-1.704	-1.639	-2.033	-2.130	-1.980
	.4300	-1.005	-1.291	-1.259	-1.028	-1.077	-1.459	-1.391	-1.231
	.5800	-1.127	-1.329	-.868	-.765	-1.154	-1.459	-.947	-.952
	.6800	-1.516	-1.828	-.755	-.770	-1.447	-1.914	-.807	-.865
Lower	.0025	.174	-1.028	-1.608	-2.413	-.683	-3.621	-2.995	-3.624
	.0050	.324	-.167	-1.174	-.854	-.096	-1.206	-2.323	-1.533
	.0100	.371	.199	-.538	-.202	.207	-.517	-1.256	-.635
	.0250	.314	.486	.193	.216	.346	.282	-.005	.106
	.0700	.474	.639	.344	.282	.495	.684	.319	.279
	.1200	.465	.600	.406	.282	.490	.580	.386	.274
	.2200	.460	.555	.344	.253	.500	.531	.357	.245
	.2800	.455	.361	.292	.230	.505	.378	.314	.221
	.4300	.441	.426	.132	.066	.466	.431	.135	.087
	.5800	.535	.393	.014	-.047	.543	.392	.010	-.010
	.7200	.643	.268	-.033	-.070	.630	.282	-.029	-.043
FLAP or AILERON									
Upper	.7312	-.307	-.371			-.274	-.368		
	.7395	-.700	-.872			-.633	-.870		
	.7532	-6.398	-9.610	-.072	-.042	-5.716	-9.664	-.090	-.063
	.7806	-1.709	-3.509	-1.236	-.789	-1.452	-3.511	-1.169	-.832
	.8354	-.620	-1.648	-.656	-.483	-.591	-1.631	-.647	-.586
	.9038	-.164	-1.375	-.481	-.385	-.173	-1.344	-.488	-.476
	.9440	.014	-.917	-.660	-.422	.034	-.914	-.642	-.519
	.9863	-.225	-.787	-.415	-.329	-.510	-.765	-.353	-.409
	.9928	.333	-.435	-.193	-.282	.322	-.421	-.198	-.322
Lower	.7312	.713	.356	-.033	-.136	.692	.364	-.029	-.087
	.7367	.709	.352	-.024	-.136	.692	.335	-.029	-.087
	.7428	.704	.389	.024	-.141	.687	.392	.005	-.087
	.7532	.704	.333	-.033	-.141	.687	.330	-.043	-.091
	.7806	.704	.287	-.005	-.103	.687	.258	-.010	-.091
	.8628	.643	.056	-.033	-.141	.620	.014	-.068	-.120
	.9313	.582	-.153	-.066	-.141	.562	-.172	-.087	-.149

TABLE 22

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60°$ $\delta_a = 00°$ $i_t = 00°$

$C_{\mu_f} = .032$ $C_{\mu,LE}$: I = .000 , C = .009 , O = .015 $C_{\mu,K}$: I = .000 , C = .000 , O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.5 °				α = 9.1 °			
WING									
Upper	.0000	.386	.266	-.462	-.026	.241	-1.415	-3.850	-3.084
	.0025	.466	-.300	-1.160	-.648	-.184	-2.064	-4.485	-3.442
	.0150	.258				-.245			
	.0250	.182	-.481	-.729	.065	-.302	-1.267	-1.869	-1.046
	.0400	.106	-.451	-.631	-.122	-.335	-1.170	-1.518	-1.000
	.0700	-.081	-.451	-.533	-.209	-.500	-.982	-1.238	-.944
	.1200	-.436	-.781	-1.013	-1.030	-.896	-1.322	-1.701	-1.772
	.1900	-1.250	-1.781	-1.764	-2.352	-1.726	-2.336	-2.387	-3.163
	.2200	-.911	-1.382	-1.502	-1.326	-1.278	-1.793	-1.972	-1.842
	.2800	-.843	-1.201	-1.258	-1.100	-1.155	-1.539	-1.612	-1.507
	.4300	-.746	-1.026	-.858	-.639	-.962	-1.230	-1.093	-.888
	.5800	-.928	-1.176	-.635	-.469	-1.113	-1.327	-.785	-.660
	.6800	-1.360	-1.652	-.551	-.430	-1.537	-1.834	-.682	-.619
Lower	.0025	.093	.412	.284	.270	.457	.101	-.266	-.335
	.0050	.140	.318	.382	.317	.406	.267	-.047	.102
	.0100	.169	.360	.382	.270	.363	.438	.154	.279
	.0250	.064	.408	.329	.261	.250	.498	.322	.307
	.0700	.314	.365	.329	.222	.387	.498	.341	.274
	.1200	.369	.365	.369	.287	.429	.450	.374	.284
	.2200	.380	.400	.302	.256	.430	.400	.313	.237
	.2800	.394	.335	.267	.213	.420	.327	.271	.205
	.4300	.373	.395	.071	.065	.396	.382	.103	.070
	.5800	.513	.373	-.067	-.035	.528	.355	-.047	-.060
	.7200	.614	.249	-.084	.009	.613	.226	-.084	-.037
FLAP or AILERON									
Upper	.7312	-.279	-.315			-.300	-.343		
	.7395	-.711	-.785			-.735	-.839		
	.7532	-6.648	-9.024	-.084	-.043	-6.890	-9.557	-.082	-.058
	.7806	-1.835	-3.347	-1.022	-.409	-1.934	-3.580	-1.070	-.721
	.8354	-.699	-1.643	-.538	-.243	-.755	-1.746	-.593	-.433
	.9038	-.212	-1.343	-.373	-.174	-.236	-1.433	-.434	-.316
	.9440	.047	-.918	-.529	-.196	.038	-.977	-.621	-.381
	.9863	-.136	-.781	-.302	-.143	-.151	-.843	-.350	-.270
	.9928	.326	-.498	-.151	-.130	.340	-.539	-.192	-.195
Lower	.7312	.699	.326	-.116	-.113	.703	.318	-.117	-.135
	.7367	.686	.305	-.093	-.113	.693	.300	-.117	-.135
	.7428	.686	.343	-.053	-.074	.693	.346	-.061	-.070
	.7532	.686	.287	-.084	-.117	.693	.281	-.093	-.144
	.7806	.686	.227	-.084	-.061	.693	.226	-.098	-.093
	.8628	.648	.030	-.084	-.061	.632	.014	-.098	-.116
	.9313	.551	-.180	-.084	-.061	.556	-.203	-.112	-.116

TABLE 22 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 45° Droop

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} = .032$ $C_{\mu,LE}$: I = .000 ,C = .009 ,O = .015 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.8 °				α = 16.5 °			
WING									
Upper	.0000	-1.270	-5.323	-10.007	-8.895	-4.191	-11.022	-18.804	-16.242
	.0025	-1.791	-5.489	-9.932	-8.720	-4.358	-10.181	-17.963	-16.079
	.0150	-1.033				-2.071			
	.0250	-.929	-2.342	-3.348	-2.625	-1.722	-3.578	-5.104	-4.567
	.0400	-.853	-2.081	-2.613	-2.152	-1.469	-3.090	-3.891	-3.524
	.0700	-.943	-1.624	-2.066	-1.915	-1.430	-2.251	-3.038	-3.024
	.1200	-1.341	-1.957	-2.438	-2.640	-1.770	-2.625	-3.246	-3.557
	.1900	-2.118	-2.876	-2.886	-4.042	-2.574	-3.313	-3.384	-4.922
	.2200	-1.625	-2.190	-2.377	-2.384	-2.483	-2.493	-2.768	-2.894
	.2800	-1.327	-1.823	-1.948	-1.896	-1.818	-2.028	-2.246	-2.254
	.4300	-1.038	-1.376	-1.250	-1.133	-1.081	-1.455	-1.393	-1.336
	.5800	-1.166	-1.438	-.896	-.844	-1.124	-1.441	-.976	-1.005
	.6800	-1.516	-1.914	-.750	-.791	-1.397	-1.862	-.801	-.923
Lower	.0025	.123	-1.238	-1.570	-2.526	-.880	-3.483	-3.668	-6.057
	.0050	.299	-.276	-1.141	-.896	-.206	-1.137	-2.824	-2.577
	.0100	.351	.105	-.514	-.194	.196	-.502	-1.550	-1.168
	.0250	.308	.448	.208	.223	.340	.313	-.090	-.038
	.0700	.455	.624	.344	.275	.521	.706	.332	.255
	.1200	.460	.500	.415	.270	.507	.600	.422	.279
	.2200	.460	.519	.349	.256	.520	.559	.393	.255
	.2800	.460	.450	.302	.223	.526	.500	.370	.236
	.4300	.436	.400	.141	.057	.502	.427	.190	.096
	.5800	.526	.357	.000	-.043	.574	.398	.066	.000
	.7200	.616	.248	-.033	-.071	.641	.303	.014	-.043
FLAP or AILERON									
Upper	.7312	-.278	-.350			-.245	-.329		
	.7395	-.689	-.854			-.606	-.811		
	.7532	-6.322	-9.508	-.083	-.065	-5.430	-9.123	-.125	-.084
	.7806	-1.744	-3.509	-1.080	-.801	-1.416	-3.279	-1.256	-.846
	.8354	-.682	-1.666	-.599	-.517	-.565	-1.450	-.663	-.610
	.9038	-.194	-1.362	-.424	-.398	-.187	-1.156	-.479	-.529
	.9440	.057	-.871	-.637	-.450	.029	-.739	-.602	-.481
	.9863	-.303	-.800	-.311	-.308	-.349	-.607	-.299	-.399
	.9928	.327	-.433	-.156	-.223	.306	-.318	-.147	-.327
Lower	.7312	.697	.343	-.052	-.128	.718	.379	.033	-.077
	.7367	.687	.324	-.052	-.128	.713	.370	.028	-.077
	.7428	.682	.376	.000	-.024	.713	.436	.085	.062
	.7532	.682	.319	-.038	-.133	.708	.370	.052	-.106
	.7806	.682	.248	-.038	-.076	.708	.313	.052	-.062
	.8628	.630	.029	-.038	-.118	.631	.085	.005	-.125
	.9313	.559	-.186	-.075	-.123	.555	-.104	-.033	-.149

TABLE 23

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 50° Droop Out'b., 45° Droop Inb'd. and Center

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .033 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .001 ,C = .003 ,O = .010

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.5°				α = 9.1°			
WING									
Upper	.0000	.354	.386	.151	.268	.315	-.954	-2.084	-1.594
	.0025	.467	.009	-.267	-.045	-.064	-1.434	-2.656	-1.822
	.0150	.279				-.155			
	.0250	.192	-.215	-.413	-.326	-.196	-.927	-1.344	-1.173
	.0400	.140	-.338	-.431	-.330	-.228	-.973	-1.177	-1.009
	.0700	-.044	-.443	-.453	-.406	-.388	-.945	-1.102	-.995
	.1200	-.389	-.763	-.875	-.835	-.767	-1.269	-1.498	-1.429
	.1900	-1.140	-1.618	-1.515	-2.214	-1.511	-2.087	-2.093	-2.758
	.2200	-.843	-1.329	-1.400	-1.299	-1.105	-1.657	-1.786	-1.630
	.2800	-.795	-1.171	-1.142	-.991	-1.014	-1.406	-1.456	-1.278
	.4300	-.703	-.995	-.822	-.620	-.845	-1.119	-.991	-.763
	.5800	-.886	-1.123	-.564	-.433	-.991	-1.242	-.674	-.530
	.6800	-1.310	-1.627	-.502	-.411	-1.393	-1.708	-.581	-.511
Lower	.0025	.083	.368	.316	.192	.489	.247	.060	.105
	.0050	.140	.303	.382	.223	.447	.347	.209	.269
	.0100	.179	.338	.378	.196	.379	.479	.316	.347
	.0250	.074	.399	.311	.196	.256	.511	.391	.342
	.0700	.310	.338	.333	.214	.411	.511	.391	.292
	.1200	.362	.355	.369	.286	.452	.500	.433	.320
	.2200	.380	.340	.338	.254	.450	.450	.377	.283
	.2800	.406	.316	.284	.219	.447	.384	.321	.260
	.4300	.384	.381	.089	.067	.420	.425	.140	.110
	.5800	.511	.351	-.031	-.031	.539	.393	.019	-.014
	.7200	.603	.237	-.062	.009	.630	.279	-.009	.018
FLAP or AILERON									
Upper	.7312	-.285	-.328			-.290	-.339		
	.7395	-.717	-.781			-.729	-.803		
	.7532	-6.575	-8.463	-.101	-.066	-6.621	-8.748	-.135	-.098
	.7806	-1.812	-3.056	-1.124	-.603	-1.813	-3.183	-1.284	-.845
	.8354	-.672	-1.469	-.547	-.295	-.671	-1.511	-.628	-.416
	.9038	-.236	-1.149	-.356	-.210	-.219	-1.192	-.437	-.288
	.9440	-.061	-.741	-.467	-.250	-.050	-.767	-.540	-.365
	.9863	-.022	-.614	-.289	-.174	-.146	-.671	-.312	-.233
	.9928	.253	-.355	-.124	-.134	.260	-.356	-.126	-.142
Lower	.7312	.699	.377	-.040	-.094	.717	.384	.023	-.046
	.7367	.694	.351	-.013	-.080	.717	.374	.047	-.027
	.7428	.694	.395	.022	.004	.712	.384	.098	.078
	.7532	.694	.346	-.027	-.094	.703	.356	.033	-.050
	.7806	.694	.281	.013	-.031	.703	.306	.056	.023
	.8628	.642	.105	-.018	-.045	.653	.119	.028	-.018
	.9313	.563	-.057	-.040	-.045	.589	-.068	-.028	-.037

TABLE 23 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 50° Droop Out'b., 45° Droop Inb'd. and Center

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .033 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .001 ,C = .003 ,O = .010

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.8°				α = 16.5°			
WING									
Upper	.0000	-1.023	-4.358	-6.849	-5.870	-3.679	-10.021	-12.256	-10.478
	.0025	-1.514	-4.488	-6.868	-5.679	-3.860	-8.951	-11.606	-9.187
	.0150	-.836				-1.791			
	.0250	-.766	-1.935	-2.677	-2.591	-1.498	-3.216	-4.446	-4.414
	.0400	-.701	-1.758	-2.224	-2.060	-1.279	-2.759	-3.682	-3.575
	.0700	-.794	-1.512	-1.953	-1.888	-1.242	-2.231	-3.129	-2.995
	.1200	-1.177	-1.814	-2.243	-2.232	-1.572	-2.481	-2.932	-2.863
	.1900	-1.817	-2.512	-2.616	-3.605	-2.060	-2.948	-3.009	-4.065
	.2200	-1.294	-1.963	-2.196	-2.079	-1.451	-2.287	-2.538	-2.372
	.2800	-1.187	-1.623	-1.771	-1.628	-1.298	-1.853	-2.057	-1.853
	.4300	-.944	-1.246	-1.168	-.967	-.977	-1.363	-1.283	-1.108
	.5800	-1.023	-1.288	-.790	-.688	-1.023	-1.363	-.875	-.788
	.6800	-1.392	-1.735	-.673	-.628	-1.344	-1.778	-.687	-.674
Lower	.0025	.215	-.898	-.906	-1.330	-.707	-3.230	-2.375	-3.273
	.0050	.355	-.107	-.598	-.344	-.116	-1.066	-1.807	-1.320
	.0100	.406	.251	-.168	.084	.228	-.387	-.909	-.448
	.0250	.360	.512	.299	.326	.381	.358	.115	.198
	.0700	.481	.656	.392	.326	.530	.717	.385	.340
	.1200	.477	.600	.439	.340	.526	.650	.447	.363
	.2200	.490	.586	.374	.312	.540	.566	.399	.325
	.2800	.491	.500	.332	.279	.553	.448	.356	.288
	.4300	.467	.456	.178	.140	.507	.448	.207	.160
	.5800	.556	.419	.061	.019	.591	.448	.082	.061
	.7200	.654	.302	.033	.028	.665	.349	.038	.042
FLAP or AILERON									
Upper	.7312	-.274	-.330			-.252	-.323		
	.7395	-.700	-.788			-.634	-.765		
	.7532	-6.317	-8.595	-.118	-.082	-5.660	-8.338	-.175	-.126
	.7806	-1.677	-3.102	-1.210	-.753	-1.456	-2.891	-1.336	-.811
	.8354	-.589	-1.428	-.593	-.433	-.484	-1.349	-.654	-.490
	.9038	-.164	-1.121	-.420	-.307	-.084	-1.047	-.442	-.358
	.9440	.014	-.693	-.547	-.353	.051	-.599	-.510	-.358
	.9863	-.257	-.665	-.299	-.233	-.200	-.571	-.245	-.241
	.9928	.294	-.293	-.117	-.158	.405	-.198	-.096	-.141
Lower	.7312	.724	.409	.051	-.023	.721	.443	.091	.028
	.7367	.719	.405	.075	-.023	.721	.439	.111	.028
	.7428	.719	.451	.112	.107	.721	.490	.163	.184
	.7532	.719	.400	.061	-.028	.721	.434	.115	.009
	.7806	.719	.326	.070	.005	.721	.368	.120	.042
	.8628	.659	.135	.033	-.023	.665	.165	.072	-.014
	.9313	.593	-.042	-.005	-.033	.619	.014	.005	-.057

TABLE 24

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 00° i_t = 00°

C_{μ_f} = .100 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.2°				α = 8.8°			
WING									
Upper	.0000	.467	-.183	-1.300	-.731	-.420	-2.828	-5.517	-3.999
	.0025	.233	-.749	-1.928	-1.286	-.925	-3.110	-5.740	-4.368
	.0150	.021				-.569			
	.0250	-.034	-.616	-1.098	-.981	-.569	-1.518	-2.263	-2.027
	.0400	-.073	-.718	-.990	-.799	-.541	-1.444	-1.886	-1.622
	.0700	-.238	-.727	-.919	-.799	-.654	-1.272	-1.658	-1.483
	.1200	-.562	-1.035	-1.242	-1.145	-.999	-1.555	-1.831	-1.704
	.1900	-1.264	-1.919	-1.780	-2.122	-1.677	-2.360	-2.036	-2.350
	.2200	-.943	-1.562	-1.538	-1.354	-1.242	-1.948	-2.154	-1.875
	.2800	-.874	-1.334	-1.309	-1.108	-1.139	-1.601	-1.849	-1.612
	.4300	-.805	-1.182	-.910	-.672	-.985	-1.323	-1.045	-.857
	.5800	-1.060	-1.365	-.641	-.472	-1.200	-1.467	-.754	-.594
	.6800	-1.575	-1.946	-.551	-.409	-1.705	-2.041	-.613	-.497
Lower	.0025	.376	.383	.147	.231	.350	-.374	-.618	-.658
	.0050	.346	.361	.295	.331	.397	.101	-.359	-.027
	.0100	.303	.459	.345	.327	.420	.379	-.036	.230
	.0250	.155	.455	.345	.281	.313	.513	.309	.331
	.0700	.350	.406	.322	.204	.429	.574	.359	.281
	.1200	.406	.312	.354	.263	.439	.550	.399	.281
	.2200	.400	.300	.282	.236	.430	.555	.331	.244
	.2800	.389	.303	.237	.190	.425	.388	.286	.211
	.4300	.380	.379	.049	.018	.420	.388	.095	.059
	.5800	.523	.348	-.107	-.095	.527	.347	-.050	-.050
	.7200	.610	.169	-.134	-.045	.630	.203	-.090	-.050
FLAP or AILERON									
Upper	.7312	.117	.107			.120	.115		
	.7395	-1.265	-1.159			-1.330	-1.171		
	.7532	-12.057	-16.700	-.095	-.074	-12.316	-17.710	-.126	-.086
	.7806	-2.484	-7.459	-1.349	-.595	-2.583	-8.003	-1.649	-.539
	.8354	-1.151	-2.888	-.623	-.322	-1.200	-3.022	-.699	-.368
	.9038	-.385	-1.687	-.376	-.218	-.411	-1.782	-.427	-.276
	.9440	-.341	-1.330	-.542	-.254	-.350	-1.444	-.540	-.258
	.9863	-.090	-1.200	-.358	-.195	.000	-1.319	-.399	-.221
	.9928	-.012	-.870	-.192	-.168	.000	-.953	-.218	-.216
Lower	.7312	.683	.290	-.210	-.213	.682	.296	-.136	-.175
	.7367	.683	.267	-.174	-.168	.682	.259	-.118	-.161
	.7428	.683	.312	-.156	-.172	.682	.300	-.104	-.133
	.7532	.683	.223	-.201	-.172	.682	.231	-.131	-.152
	.7806	.683	.156	-.161	-.122	.696	.157	-.113	-.101
	.8628	.640	-.071	-.161	-.131	.649	-.101	-.131	-.138
	.9313	.571	-.303	-.161	-.118	.593	-.374	-.163	-.138

TABLE 24 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60°$ $\delta_a = 00°$ $i_t = 00°$

$C_{\mu_f} =$ .100 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.5°				α = 16.5°			
WING									
Upper	.0000	-2.321	-7.362	-11.712	-6.864	-5.695	-12.306	-2.748	-2.429
	.0025	-2.737	-7.089	-11.220	-6.479	-5.564	-9.748	-2.706	-2.209
	.0150	-1.350				-2.429			
	.0250	-1.177	-2.650	-3.588	-3.215	-1.962	-3.506	-2.646	-2.153
	.0400	-1.004	-2.320	-2.975	-2.518	-1.630	-2.919	-2.613	-2.116
	.0700	-1.009	-1.899	-2.574	-2.067	-1.495	-2.269	-2.613	-2.074
	.1200	-1.340	-2.138	-2.291	-1.768	-1.747	-2.336	-2.409	-1.752
	.1900	-1.952	-2.760	-2.046	-1.615	-2.289	-2.530	-1.344	-1.177
	.2200	-1.485	-2.564	-2.466	-1.663	-2.111	-2.487	-1.646	-1.111
	.2800	-1.284	-1.927	-2.320	-1.586	-1.658	-2.227	-1.553	-1.111
	.4300	-1.027	-1.516	-1.551	-1.355	-1.088	-1.682	-1.390	-1.069
	.5800	-1.210	-1.616	-1.094	-1.153	-1.219	-1.464	-1.288	-1.032
	.6800	-1.686	-2.191	-.881	-1.009	-1.621	-1.469	-1.223	-1.037
Lower	.0025	-.191	-2.100	-1.919	-2.124	-1.434	-4.127	-.772	-1.074
	.0050	.163	-.607	-1.452	-.735	-.527	-1.445	-.590	-.308
	.0100	.345	-.100	-.683	-.124	.060	-.649	-.209	.074
	.0250	.378	.401	.165	.249	.341	.251	.288	.299
	.0700	.495	.669	.386	.293	.481	.710	.418	.327
	.1200	.490	.600	.448	.293	.518	.600	.465	.322
	.2200	.490	.545	.382	.254	.530	.563	.404	.284
	.2800	.485	.425	.344	.216	.537	.526	.367	.256
	.4300	.481	.420	.155	.057	.532	.450	.199	.084
	.5800	.565	.358	-.004	-.076	.593	.412	.032	-.037
	.7200	.649	.215	-.075	-.149	.677	.322	-.088	-.121
FLAP or AILERON									
Upper	.7312	.122	.115			.120	.122		
	.7395	-1.282	-1.189			-1.201	-.489		
	.7532	-11.959	-17.581	-.190	-.143	-11.249	-8.696	-.143	-.127
	.7806	-2.387	-8.147	-2.122	-.975	-2.149	-3.748	-1.525	-1.079
	.8354	-1.060	-3.085	-1.018	-.894	-.939	-1.379	-1.069	-.976
	.9038	-.327	-1.837	-.674	-.817	-.252	-.815	-.916	-.906
	.9440	-.270	-1.473	-.532	-.778	-.214	-.706	-.437	-.784
	.9863	.046	-1.368	-.598	-.754	.056	-.672	-.795	-.826
	.9928	.046	-.975	-.424	-.697	.079	-.630	-.818	-.831
Lower	.7312	.705	.306	-.099	-.221	.691	.398	-.144	-.191
	.7367	.705	.277	-.103	-.221	.691	.369	-.130	-.191
	.7428	.705	.330	-.080	-.144	.719	.421	-.130	-.112
	.7532	.705	.253	-.113	-.245	.710	.355	-.158	-.219
	.7806	.710	.196	-.084	-.177	.710	.326	-.148	-.172
	.8628	.663	-.110	-.155	-.278	.654	.146	-.232	-.275
	.9313	.607	-.368	-.231	-.355	.602	-.047	-.353	-.378

TABLE 25

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60°$ $\delta_a = 10°$ $i_t = 00°$

C_{μ_f} = .102 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.1°				α = 8.7°			
WING									
Upper	.0000	.375	-.528	-2.477	-1.756	-.542	-3.519	-7.393	-6.057
	.0025	.107	-1.098	-3.076	-2.896	-1.139	-3.781	-7.482	-7.465
	.0150	-.103				-.716			
	.0250	-.154	-.792	-1.545	-1.331	-.706	-1.820	-2.866	-2.718
	.0400	-.197	-.886	-1.362	-1.130	-.666	-1.723	-2.403	-2.179
	.0700	-.375	-.886	-1.246	-1.088	-.771	-1.504	-2.078	-1.941
	.1200	-.736	-1.202	-1.560	-1.434	-1.159	-1.805	-2.236	-2.126
	.1900	-1.502	-2.145	-1.999	-2.485	-1.900	-2.674	-2.379	-2.669
	.2200	-1.145	-1.740	-1.874	-1.686	-1.437	-2.247	-2.679	-2.436
	.2800	-1.074	-1.485	-1.598	-1.396	-1.293	-1.834	-2.310	-2.208
	.4300	-.985	-1.311	-1.183	-.939	-1.134	-1.529	-1.369	-1.198
	.5800	-1.253	-1.504	-.956	-.756	-1.358	-1.674	-1.078	-.927
	.6800	-1.807	-2.131	-.937	-.761	-1.895	-2.286	-.999	-.864
Lower	.0025	.375	.287	-.038	.060	.328	-.582	-1.014	-1.456
	.0050	.328	.330	.115	.238	.393	-.024	-.665	-.417
	.0100	.286	.462	.227	.289	.417	.315	-.246	.009
	.0250	.150	.457	.318	.284	.333	.465	.236	.252
	.0700	.323	.429	.294	.205	.422	.533	.334	.247
	.1200	.384	.450	.347	.266	.422	.530	.399	.242
	.2200	.370	.514	.280	.210	.410	.533	.330	.208
	.2800	.361	.440	.217	.177	.398	.450	.295	.169
	.4300	.347	.382	.057	.032	.398	.388	.108	.033
	.5800	.516	.348	-.062	-.051	.537	.349	-.014	-.072
	.7200	.619	.160	-.062	-.060	.631	.150	-.068	-.106
FLAP or AILERON									
Upper	.7312	-.115	-.122			-.125	-.129		
	.7395	-14.020	-12.025			-14.256	-12.406		
	.7532	-12.497	-17.328	-.113	-.101	-12.615	-17.950	-.167	-.135
	.7806	-2.769	-7.616	-1.212	-.939	-2.795	-7.853	-1.857	-1.092
	.8354	-1.337	-2.721	-.695	-.560	-1.323	-2.800	-.886	-.669
	.9038	-.483	-1.669	-.497	-.401	-.477	-1.757	-.586	-.451
	.9440	-.427	-1.660	-.565	-.397	-.422	-1.742	-.596	-.451
	.9863	-.215	-1.754	-.376	-.308	-.363	-1.878	-.556	-.436
	.9928	-.065	-1.145	-.255	-.233	-.079	-1.194	-.334	-.383
Lower	.7312	.675	.282	.188	.214	.701	.266	.197	.116
	.7367	.675	.249	.144	.135	.696	.223	.172	.033
	.7428	.675	.282	.115	.112	.696	.281	.142	.106
	.7532	.675	.231	.028	.042	.696	.218	.059	-.029
	.7806	.690	.165	-.019	.004	.696	.145	.009	-.048
	.8628	.647	-.108	-.086	-.065	.666	-.116	-.068	-.126
	.9313	.586	-.396	-.125	-.112	.606	-.470	-.142	-.179

TABLE 25 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60^\circ$ $\delta_a = 10^\circ$ $i_t = 00^\circ$

$C_{\mu_f} = .102$ $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.4°				α = 16.5°			
WING									
Upper	.0000	-2.934	-8.483	-14.299	-9.952	-6.186	-13.283	-2.974	-2.572
	.0025	-3.348	-8.076	-13.615	-13.046	-6.113	-10.253	-2.929	-2.945
	.0150	-1.686				-2.735			
	.0250	-1.444	-3.025	-4.225	-4.920	-2.222	-3.865	-2.868	-2.268
	.0400	-1.252	-2.633	-3.560	-4.047	-1.885	-3.205	-2.858	-2.253
	.0700	-1.252	-2.169	-3.100	-3.273	-1.756	-2.500	-2.858	-2.228
	.1200	-1.605	-2.417	-2.730	-2.552	-2.020	-2.575	-2.651	-1.935
	.1900	-2.257	-3.128	-2.415	-2.768	-2.647	-2.640	-1.373	-1.412
	.2200	-1.757	-2.927	-2.895	-2.352	-2.388	-2.545	-1.823	-1.338
	.2800	-1.520	-2.221	-2.655	-2.084	-1.885	-2.305	-1.747	-1.338
	.4300	-1.232	-1.716	-1.690	-1.578	-1.331	-1.825	-1.631	-1.278
	.5800	-1.419	-1.829	-1.275	-1.278	-1.476	-1.665	-1.530	-1.198
	.6800	-1.954	-2.417	-1.115	-1.152	-1.916	-1.665	-1.464	-1.213
Lower	.0025	-.368	-2.571	-2.525	-3.131	-1.626	-4.640	-.919	-1.223
	.0050	.010	-.788	-1.930	-1.252	-.621	-1.675	-.691	-.412
	.0100	.247	-.206	-1.015	-.431	.025	-.790	-.318	-.019
	.0250	.318	.386	.065	.152	.300	.205	.232	.223
	.0700	.459	.675	.345	.252	.440	.695	.383	.258
	.1200	.444	.600	.395	.252	.492	.650	.409	.258
	.2200	.450	.561	.320	.205	.500	.560	.368	.223
	.2800	.459	.450	.295	.178	.507	.500	.323	.199
	.4300	.459	.412	.120	.057	.502	.435	.156	.034
	.5800	.540	.365	.000	-.063	.580	.390	.000	-.074
	.7200	.631	.221	-.085	-.142	.673	.290	-.146	-.159
FLAP or AILERON									
Upper	.7312	-.131	-.145			-.138	-.133		
	.7395	-15.057	-13.183			-13.377	-5.685		
	.7532	-13.283	-18.567	-.150	-.137	-11.910	-9.415	-.228	-.172
	.7806	-2.716	-7.772	-1.640	-1.168	-2.538	-3.920	-2.025	-1.278
	.8354	-1.292	-2.783	-.980	-.978	-1.181	-1.585	-1.307	-1.124
	.9038	-.459	-1.716	-.785	-.952	-.373	-.905	-1.090	-1.039
	.9440	-.409	-1.623	-.525	-.884	-.347	-.730	-.850	-.935
	.9863	-.474	-1.762	-.735	-.889	-.885	-.800	-.969	-1.004
	.9928	-.050	-1.077	-.590	-.805	-.005	-.660	-.969	-1.004
Lower	.7312	.691	.324	.195	.094	.699	.390	.156	.079
	.7367	.691	.288	.155	.021	.699	.360	.111	-.014
	.7428	.691	.350	.130	.089	.699	.415	.080	.044
	.7532	.691	.278	.040	-.047	.699	.360	-.015	-.089
	.7806	.691	.206	-.010	-.094	.699	.315	-.045	-.124
	.8628	.651	-.082	-.130	-.231	.668	.155	-.191	-.273
	.9313	.616	-.386	-.230	-.326	.590	-.055	-.373	-.407

TABLE 26

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 20° i_t = 00°

C_{μ_f} = .101 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.1°				α = 8.7°			
WING									
Upper	.0000	.423	-.574	-2.476	-2.200	-.652	-3.418	-7.692	-6.815
	.0025	.147	-1.120	-3.091	-3.223	-1.188	-3.721	-7.692	-7.499
	.0150	-.059				-.753			
	.0250	-.119	-.823	-1.532	-1.532	-.725	-1.798	-2.848	-2.995
	.0400	-.165	-.916	-1.334	-1.280	-.681	-1.721	-2.382	-2.383
	.0700	-.350	-.916	-1.224	-1.205	-.778	-1.509	-2.075	-2.121
	.1200	-.709	-1.235	-1.550	-1.555	-1.179	-1.812	-2.207	-2.267
	.1900	-1.474	-2.171	-2.054	-2.639	-1.889	-2.692	-2.367	-2.898
	.2200	-1.128	-1.763	-1.876	-1.798	-1.435	-2.250	-2.589	-2.641
	.2800	-1.055	-1.504	-1.619	-1.504	-1.304	-1.851	-2.231	-2.383
	.4300	-.986	-1.351	-1.224	-1.046	-1.121	-1.543	-1.443	-1.315
	.5800	-1.230	-1.555	-1.073	-.883	-1.357	-1.702	-1.217	-1.097
	.6800	-1.797	-2.175	-1.151	-.939	-1.879	-2.303	-1.245	-1.097
Lower	.0025	.368	.263	-.050	-.135	.295	-.582	-1.132	-1.942
	.0050	.327	.300	.128	.158	.367	-.010	-.750	-.680
	.0100	.276	.449	.224	.266	.401	.288	-.311	-.112
	.0250	.142	.449	.334	.266	.304	.452	.255	.228
	.0700	.294	.421	.325	.210	.415	.558	.358	.223
	.1200	.382	.450	.366	.252	.425	.550	.401	.223
	.2200	.380	.504	.311	.214	.430	.548	.344	.194
	.2800	.382	.420	.270	.177	.415	.450	.283	.155
	.4300	.368	.379	.123	.046	.415	.399	.127	.044
	.5800	.520	.361	.032	-.023	.541	.356	.071	-.044
	.7200	.622	.185	.064	-.023	.628	.207	.075	-.044
FLAP or AILERON									
Upper	.7312	-.039	-.043			-.046	-.049		
	.7395	-14.160	-11.871			-14.532	-12.295		
	.7532	-12.525	-17.482	-.272	-.231	-12.766	-17.828	-.251	-.229
	.7806	-2.741	-7.647	-2.183	-1.270	-2.748	-7.643	-2.009	-1.437
	.8354	-1.308	-2.448	-.931	-.733	-1.328	-2.461	-.910	-.854
	.9038	-.465	-1.272	-.619	-.635	-.473	-1.288	-.618	-.660
	.9440	-.410	-1.652	-.449	-.574	-.415	-1.610	-.476	-.578
	.9863	-.082	-1.870	-.678	-.630	-.647	-1.851	-.608	-.578
	.9928	-.050	-1.268	-.463	-.551	-.053	-1.216	-.391	-.582
Lower	.7312	.686	.259	.192	.074	.700	.308	.241	.068
	.7367	.681	.259	.252	.163	.696	.269	.307	.131
	.7428	.677	.259	.252	.238	.696	.327	.307	.252
	.7532	.677	.222	.206	.126	.696	.245	.231	.097
	.7806	.677	.162	.146	.065	.691	.212	.174	.049
	.8628	.658	-.074	.009	-.065	.662	-.043	.038	-.092
	.9313	.599	-.407	-.119	-.158	.618	-.409	-.080	-.184

TABLE 26 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60 ° δ_a = 20 ° i_t = 00 °

C_{μ_f} = .101 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	C_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.4 °				α = 16.5 °			
WING									
Upper	.0000	-2.517	-8.339	-14.884	-6.901	-5.700	-12.597	-2.864	-2.703
	.0025	-2.995	-7.984	-14.147	-12.549	-5.700	-9.879	-2.813	-2.906
	.0150	-1.517				-2.568			
	.0250	-1.320	-3.025	-4.348	-4.268	-2.112	-3.631	-2.747	-2.213
	.0400	-1.167	-2.659	-3.731	-3.886	-1.777	-3.009	-2.710	-2.198
	.0700	-1.172	-2.196	-3.284	-3.324	-1.655	-2.330	-2.710	-2.143
	.1200	-1.522	-2.469	-2.761	-2.093	-1.954	-2.402	-2.504	-1.935
	.1900	-2.197	-3.139	-2.338	-1.763	-2.538	-2.502	-1.275	-1.341
	.2200	-1.704	-2.912	-2.881	-1.737	-2.299	-2.502	-1.701	-1.307
	.2800	-1.483	-2.196	-2.721	-1.721	-1.782	-2.248	-1.649	-1.307
	.4300	-1.202	-1.752	-1.751	-1.613	-1.239	-1.741	-1.518	-1.262
	.5800	-1.384	-1.861	-1.537	-1.309	-1.391	-1.588	-1.388	-1.173
	.6800	-1.901	-2.458	-1.393	-1.206	-1.843	-1.588	-1.318	-1.178
Lower	.0025	-.261	-2.515	-2.696	-2.433	-1.411	-4.133	-.846	-1.183
	.0050	.103	-.747	-2.050	-.943	-.508	-1.402	-.631	-.386
	.0100	.291	-.191	-1.085	-.294	.076	-.598	-.257	.005
	.0250	.320	.392	.055	.160	.325	.277	.257	.277
	.0700	.453	.675	.353	.263	.457	.722	.411	.277
	.1200	.448	.630	.413	.263	.492	.650	.439	.277
	.2200	.450	.562	.368	.216	.500	.565	.392	.248
	.2800	.438	.480	.308	.186	.508	.450	.355	.223
	.4300	.438	.407	.149	.046	.503	.450	.220	.089
	.5800	.552	.381	.035	-.062	.579	.407	.084	-.005
	.7200	.626	.263	-.005	-.072	.660	.340	.014	-.010
FLAP or AILERON									
Upper	.7312	-.045	-.057			-.049	-.053		
	.7395	-14.967	-12.689			-13.320	-6.698		
	.7532	-13.080	-18.059	-.205	-.164	-11.716	-9.391	-.191	-.170
	.7806	-2.645	-7.664	-2.060	-1.170	-2.401	-3.406	-1.663	-1.297
	.8354	-1.232	-2.500	-1.303	-1.087	-1.112	-1.378	-1.121	-1.094
	.9038	-.424	-1.392	-1.159	-1.098	-.355	-.756	-.962	-1.025
	.9440	-.374	-1.613	-.557	-1.067	-.325	-.636	-.374	-.792
	.9863	-.616	-1.814	-1.119	-1.072	-.985	-.808	-.855	-.955
	.9928	-.034	-1.232	-.945	-1.031	.010	-.445	-.832	-.936
Lower	.7312	.685	.350	.164	.005	.690	.440	.173	.069
	.7367	.685	.345	.239	.093	.685	.435	.243	.173
	.7428	.685	.387	.284	.289	.685	.483	.280	.292
	.7532	.680	.309	.204	.046	.685	.431	.196	.124
	.7806	.680	.258	.124	-.015	.685	.402	.145	.064
	.8628	.650	-.005	-.075	-.206	.650	.273	-.042	-.114
	.9313	.606	-.366	-.269	-.371	.579	.086	-.229	-.282

TABLE 27

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f= 60 ° δ_a= 30 ° i_t= 00 °

C_{μ_f}= .101 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α =5.1 °				α = 8.7 °			
WING									
Upper	.0000	.394	-.662	-3.046	-2.872	-.703	-3.421	-8.066	-7.858
	.0025	.131	-1.180	-3.560	-3.682	-1.231	-3.669	-7.985	-8.177
	.0150	-.089				-.773			
	.0250	-.146	-.838	-1.653	-1.730	-.736	-1.789	-2.929	-3.135
	.0400	-.174	-.910	-1.426	-1.450	-.660	-1.670	-2.426	-2.502
	.0700	-.361	-.896	-1.296	-1.332	-.788	-1.473	-2.118	-2.236
	.1200	-.732	-1.207	-1.606	-1.654	-1.151	-1.789	-2.242	-2.328
	.1900	-1.507	-2.090	-2.083	-2.692	-1.872	-2.622	-2.379	-2.937
	.2200	-1.145	-1.739	-1.926	-1.896	-1.410	-2.201	-2.635	-2.705
	.2800	-1.075	-1.477	-1.666	-1.602	-1.283	-1.804	-2.275	-2.391
	.4300	-.990	-1.306	-1.236	-1.109	-1.113	-1.512	-1.445	-1.372
	.5800	-1.267	-1.518	-1.069	-.943	-1.339	-1.674	-1.223	-1.116
	.6800	-1.821	-2.103	-1.092	-1.000	-1.877	-2.272	-1.237	-1.116
Lower	.0025	.361	.261	-.153	-.308	.283	-.555	-1.189	-2.154
	.0050	.319	.297	.042	.085	.358	-.005	-.834	-.845
	.0100	.272	.450	.194	.261	.396	.321	-.336	-.198
	.0250	.131	.459	.319	.303	.302	.478	.227	.203
	.0700	.310	.432	.338	.251	.415	.584	.346	.261
	.1200	.371	.450	.380	.289	.429	.570	.417	.261
	.2200	.370	.513	.329	.227	.420	.560	.351	.213
	.2800	.366	.450	.282	.190	.406	.500	.303	.184
	.4300	.361	.387	.143	.081	.406	.416	.171	.072
	.5800	.516	.378	.093	.009	.538	.402	.104	-.005
	.7200	.601	.248	.120	.076	.623	.268	.118	.034
FLAP or AILERON									
Upper	.7312	-.091	-.093			-.093	-.098		
	.7395	-14.629	-12.217	-.141	-.111	-14.784	-12.161	-.140	-.122
	.7532	-12.738	-17.254	-.218	-.203	-12.765	-17.383	-.212	-.200
	.7806	-2.760	-6.815	-1.741	-1.332	-2.683	-7.233	-1.592	-1.319
	.8354	-1.347	-2.171	-1.042	-.967	-1.297	-2.287	-1.019	-1.014
	.9038	-.488	-1.081	-.921	-.938	-.481	-1.191	-.919	-.961
	.9440	-.427	-1.464	-.850	-.725	-.410	-1.536	-.850	-.773
	.9863	-.667	-1.522	-1.023	-.990	-.962	-1.650	-.976	-.990
	.9928	-.047	-1.081	-.838	-.862	-.047	-1.167	-.829	-.898
Lower	.7312	.685	.356	.153	.100	.674	.368	.156	.053
	.7367	.676	.333	.199	.137	.689	.340	.199	.106
	.7428	.681	.365	.241	.246	.684	.383	.265	.251
	.7532	.681	.297	.259	.190	.703	.325	.261	.174
	.7806	.699	.257	.213	.152	.707	.292	.204	.111
	.8628	.652	.090	.032	-.009	.646	.096	.043	-.039
	.9313	.587	-.203	-.130	-.180	.594	-.230	-.128	-.188

TABLE 27 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f= 60° δ_a= 30° i_t= 00°

C_{μ_f}= .101 $C_{\mu,LE}$: I =.000 ,C =.000 ,O =.000 $C_{\mu,K}$: I =.000 ,C =.000 ,O =.000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α =12.4°				α = 16.5°			
WING									
Upper	.0000	-2.884	-8.820	-13.180	-6.722	-5.973	-12.512	-2.761	-2.406
	.0025	-3.338	-8.360	-12.372	11.332	-5.978	-10.100	-2.711	-2.651
	.0150	-1.672				-2.675			
	.0250	-1.449	-3.125	-4.759	-4.192	-2.193	-3.659	-2.657	-1.838
	.0400	-1.278	-2.720	-4.136	-4.040	-1.848	-3.059	-2.632	-1.857
	.0700	-1.283	-2.235	-3.568	-3.707	-1.717	-2.361	-2.627	-1.843
	.1200	-1.651	-2.470	-2.693	-2.368	-2.005	-2.410	-2.463	-1.730
	.1900	-2.293	-3.105	-2.352	-1.768	-2.654	-2.463	-1.284	-1.250
	.2200	-1.793	-2.945	-2.749	-1.717	-2.377	-2.366	-1.746	-1.196
	.2800	-1.555	-2.250	-2.452	-1.722	-1.848	-2.176	-1.696	-1.206
	.4300	-1.257	-1.755	-1.693	-1.540	-1.304	-1.741	-1.567	-1.157
	.5800	-1.439	-1.835	-1.492	-1.252	-1.429	-1.580	-1.393	-1.044
	.6800	-1.959	-2.390	-1.357	-1.237	-1.921	-1.619	-1.328	-1.044
Lower	.0025	-.369	-2.715	-2.623	-2.358	-1.502	-4.288	-.821	-.961
	.0050	.035	-.855	-2.030	-.929	-.513	-1.507	-.602	-.279
	.0100	.263	-.250	-1.050	-.283	.089	-.698	-.244	.059
	.0250	.323	.340	.085	.182	.356	.224	.264	.284
	.0700	.465	.650	.382	.283	.492	.702	.408	.309
	.1200	.460	.600	.437	.283	.518	.600	.453	.309
	.2200	.450	.545	.382	.232	.520	.541	.403	.260
	.2800	.455	.480	.347	.212	.524	.473	.363	.240
	.4300	.455	.400	.216	.086	.524	.434	.239	.127
	.5800	.561	.380	.126	.010	.607	.385	.149	.059
	.7200	.636	.290	.121	.005	.681	.317	.114	.083
FLAP or AILERON									
Upper	.7312	-.098	-.107			-.100	-.095		
	.7395	-14.756	-12.271	-.171	-.131	-13.448	-5.346	-.177	-.133
	.7532	-12.670	-17.488	-.182	-.156	-11.718	-8.815	-.167	-.138
	.7806	-2.682	-7.260	-1.904	-1.202	-2.497	-3.619	-1.627	-1.162
	.8354	-1.278	-2.255	-1.216	-1.081	-1.157	-1.541	-1.194	-.956
	.9038	-.470	-1.240	-1.126	-1.081	-.366	-.937	-1.005	-.882
	.9440	-.414	-1.550	-1.000	-.975	-.356	-.790	-.363	-.657
	.9863	-.990	-1.745	-1.106	-1.035	-1.010	-.849	-.900	-.833
	.9928	-.015	-1.295	-.960	-1.000	.031	-.659	-.900	-.833
Lower	.7312	.692	.380	.161	.015	.707	.395	.174	.113
	.7367	.692	.345	.211	.061	.733	.390	.194	.167
	.7428	.692	.400	.281	.263	.728	.424	.274	.299
	.7532	.692	.340	.281	.111	.722	.380	.274	.225
	.7806	.692	.295	.216	.061	.722	.332	.214	.172
	.8628	.672	.085	.015	-.121	.691	.190	.050	-.015
	.9313	.616	-.245	-.196	-.303	.623	-.034	-.194	-.186

TABLE 28

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60°$ $\delta_a = 40°$ $i_t = 00°$

$C_{\mu_f} =$ .102 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.1 °				α = 8.7 °			
WING									
Upper	.0000	.389	-.689	-2.881	-2.613	-.471	-3.527	-7.790	-7.470
	.0025	.084	-1.182	-3.337	-3.320	-.969	-3.727	-7.660	-7.890
	.0150	-.080				-.587			
	.0250	-.115	-.822	-1.517	-1.560	-.592	-1.700	-2.732	-3.027
	.0400	-.137	-.875	-1.298	-1.284	-.569	-1.577	-2.259	-2.402
	.0700	-.327	-.880	-1.180	-1.187	-.650	-1.445	-1.973	-2.114
	.1200	-.659	-1.182	-1.443	-1.484	-1.013	-1.641	-2.076	-2.169
	.1900	-1.363	-2.049	-1.886	-2.484	-1.677	-2.418	-2.187	-2.762
	.2200	-1.031	-1.689	-1.741	-1.729	-1.264	-2.004	-2.446	-2.461
	.2800	-.956	-1.431	-1.504	-1.435	-1.139	-1.650	-2.107	-2.164
	.4300	-.880	-1.253	-1.114	-.991	-.978	-1.373	-1.303	-1.247
	.5800	-1.115	-1.449	-.965	-.858	-1.197	-1.541	-1.076	-1.046
	.6800	-1.650	-2.004	-1.004	-.920	-1.699	-2.036	-1.080	-1.041
Lower	.0025	.403	.284	-.092	-.253	.354	-.618	-1.138	-2.365
	.0050	.376	.338	.096	.133	.413	-.032	-.795	-.868
	.0100	.327	.467	.237	.298	.430	.327	-.286	-.210
	.0250	.199	.480	.381	.356	.359	.500	.272	.247
	.0700	.358	.467	.381	.316	.462	.604	.411	.329
	.1200	.403	.500	.434	.316	.462	.600	.455	.324
	.2200	.400	.538	.381	.280	.460	.568	.415	.292
	.2800	.398	.500	.342	.262	.457	.445	.379	.265
	.4300	.398	.436	.224	.160	.453	.436	.263	.164
	.5800	.526	.418	.193	.093	.556	.436	.214	.100
	.7200	.615	.329	.219	.160	.641	.336	.210	.128
FLAP or AILERON									
Upper	.7312	-.074	-.083	-.067	-.072	-.076	-.079	-.070	-.077
	.7395	-13.531	-11.158	-.155	-.124	-13.780	-11.246	-.155	-.123
	.7532	-11.798	-15.900	-.168	-.159	-11.935	-16.113	-.169	-.157
	.7806	-2.508	-6.586	-1.456	-1.138	-2.475	-6.668	-1.469	-1.068
	.8354	-1.208	-1.969	-.943	-.902	-1.170	-1.927	-.937	-.936
	.9038	-.416	-1.102	-.912	-.947	-.395	-1.104	-.937	-1.023
	.9440	-.354	-1.191	-.303	-.702	-.327	-1.168	-.335	-.927
	.9863	-.668	-1.315	-.908	-.929	-.951	-1.332	-.911	-.982
	.9928	.013	-1.133	-.855	-.915	.013	-1.150	-.835	-.945
Lower	.7312	.690	.409	.254	.151	.708	.423	.254	.128
	.7367	.690	.387	.254	.173	.708	.418	.263	.128
	.7428	.686	.422	.285	.267	.708	.445	.308	.274
	.7532	.686	.382	.320	.244	.708	.405	.308	.196
	.7806	.686	.333	.298	.222	.704	.359	.286	.155
	.8628	.664	.191	.140	.084	.673	.227	.152	.027
	.9313	.597	-.049	-.039	-.089	.610	-.050	-.013	-.123

TABLE 28 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60^\circ$ $\delta_a = 40^\circ$ $i_t = 00^\circ$

$C_{\mu_f} =$ .102 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .000 ,O = .000

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.4°				α = 16.5°			
WING									
Upper	.0000	-2.286	-8.258	-10.517	-8.328	-5.565	-11.662	-2.577	-2.106
	.0025	-2.760	-7.791	-9.343	-9.577	-5.494	-9.379	-2.529	-2.216
	.0150	-1.389				-2.438			
	.0250	-1.188	-2.867	-4.384	-3.754	-1.962	-3.370	-2.480	-1.696
	.0400	-1.042	-2.495	-3.849	-3.615	-1.636	-2.794	-2.446	-1.700
	.0700	-1.047	-2.047	-3.291	-3.365	-1.514	-2.146	-2.432	-1.691
	.1200	-1.404	-2.287	-2.256	-2.328	-1.787	-2.183	-2.267	-1.622
	.1900	-2.028	-2.839	-1.836	-1.666	-2.349	-2.228	-1.204	-1.106
	.2200	-1.549	-2.660	-2.106	-1.569	-2.099	-2.228	-1.626	-1.101
	.2800	-1.338	-2.033	-1.845	-1.541	-1.627	-2.000	-1.587	-1.106
	.4300	-1.066	-1.575	-1.380	-1.287	-1.104	-1.566	-1.432	-1.037
	.5800	-1.281	-1.665	-1.243	-1.083	-1.221	-1.429	-1.306	-.922
	.6800	-1.741	-2.202	-1.110	-1.032	-1.660	-1.452	-1.247	-.926
Lower	.0025	-.155	-2.495	-2.035	-2.055	-1.372	-3.918	-.738	-.857
	.0050	.207	-.755	-1.553	-.875	-.495	-1.347	-.563	-.240
	.0100	.371	-.179	-.774	-.268	.099	-.566	-.194	.088
	.0250	.390	.387	.190	.241	.363	.274	.272	.336
	.0700	.512	.684	.438	.347	.486	.726	.403	.373
	.1200	.512	.640	.478	.361	.505	.650	.466	.369
	.2200	.500	.585	.442	.310	.500	.584	.422	.327
	.2800	.493	.457	.407	.296	.538	.530	.393	.318
	.4300	.512	.453	.296	.199	.538	.475	.262	.226
	.5800	.601	.439	.234	.139	.608	.443	.204	.157
	.7200	.681	.358	.230	.199	.679	.379	.209	.217
FLAP or AILERON									
Upper	.7312	-.076	-.084	-.063	-.064	-.080	-.076	-.075	-.065
	.7395	-13.525	-11.179	-.142	-.127	-12.653	-4.895	-.170	-.130
	.7532	-11.637	-15.928	-.139	-.129	-10.994	-8.146	-.154	-.122
	.7806	-2.441	-6.758	-1.504	-1.231	-2.198	-3.319	-1.500	-1.087
	.8354	-1.094	-1.943	-.973	-.930	-.976	-1.388	-1.044	-.839
	.9038	-.343	-1.005	-.920	-.907	-.283	-.826	-.908	-.756
	.9440	-.305	-1.085	-.850	-.800	-.278	-.721	-.850	-.470
	.9863	-.981	-1.320	-.823	-.852	-.920	-.840	-.903	-.737
	.9928	.061	-1.089	-.748	-.792	.042	-.626	-.849	-.733
Lower	.7312	.728	.429	.257	.218	.703	.447	.238	.226
	.7367	.728	.424	.283	.227	.707	.420	.243	.240
	.7428	.723	.453	.310	.347	.712	.466	.267	.336
	.7532	.723	.415	.332	.273	.712	.416	.320	.295
	.7806	.723	.373	.292	.264	.712	.388	.291	.267
	.8628	.676	.212	.146	.106	.670	.260	.121	.138
	.9313	.634	-.090	-.040	-.056	.599	.027	-.078	-.046

TABLE 29

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60°$ $\delta_a = 00°$ $i_t = 00°$

$C_{\mu_f} =$ .100 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.2°				α = 8.8°			
WING									
Upper	.0000	.422	-.391	-1.854	-1.136	-.429	-3.429	-6.621	-5.726
	.0025	.183	-.935	-2.532	-2.313	-.975	-3.663	-6.856	-6.621
	.0150	-.055				-.648			
	.0250	-.110	-.732	-1.354	-1.145	-.653	-1.724	-2.705	-2.564
	.0400	-.160	-.829	-1.214	-.968	-.624	-1.626	-2.293	-2.085
	.0700	-.339	-.843	-1.130	-.945	-.751	-1.420	-2.019	-1.889
	.1200	-.715	-1.179	-1.481	-1.331	-1.136	-1.729	-2.239	-2.152
	.1900	-1.467	-1.603	-1.859	-2.436	-1.873	-2.070	-2.440	-3.205
	.2200	-1.119	-1.594	-1.644	-1.504	-1.414	-1.892	-2.112	-1.999
	.2800	-1.036	-1.368	-1.350	-1.077	-1.282	-1.617	-1.730	-1.454
	.4300	-.944	-1.225	-1.004	-.686	-1.102	-1.364	-1.239	-.952
	.5800	-1.206	-1.469	-.691	-.509	-1.336	-1.584	-.857	-.712
	.6800	-1.779	-2.055	-.644	-.504	-1.892	-2.154	-.759	-.684
Lower	.0025	.357	.340	.070	.159	.351	-.612	-.823	-1.066
	.0050	.311	.317	.219	.286	.385	-.005	-.519	-.291
	.0100	.275	.446	.284	.299	.390	.299	-.142	.100
	.0250	.123	.446	.327	.259	.312	.453	.269	.277
	.0700	.321	.387	.284	.195	.414	.565	.347	.234
	.1200	.380	.450	.355	.245	.414	.187	.387	.272
	.2200	.370	.497	.261	.209	.410	.533	.313	.229
	.2800	.366	.336	.219	.159	.409	.383	.240	.186
	.4300	.362	.359	.023	.000	.390	.383	.053	.014
	.5800	.522	.322	-.158	-.109	.517	.313	-.117	-.100
	.7200	.623	.138	-.196	-.072	.629	.150	-.151	-.100
FLAP or AILERON									
Upper	.7312	-.043	-.053			-.052	-.053		
	.7395	-1.421	-1.289			-1.468	-1.307		
	.7532	-12.366	-17.893			-12.586	-18.815		
	.7806	-2.669	-7.981	-1.153	-.677	-2.663	-8.213	-1.166	-.861
	.8354	-1.458	-3.220	-.588	-.359	-1.468	-3.270	-.637	-.440
	.9038	-.600	-2.009	-.392	-.259	-.609	-2.042	-.436	-.354
	.9440	-.463	-1.686	-.612	-.345	-.487	-1.715	-.715	-.492
	.9863	-.288	-1.806	-.532	-.295	-.375	-1.827	-.553	-.401
	.9928	-.077	-1.170	-.266	-.209	-.078	-1.173	-.279	-.267
Lower	.7312	.688	.239	-.289	-.236	.692	.234	-.240	-.239
	.7367	.688	.211	-.247	-.209	.687	.192	-.215	-.205
	.7428	.688	.244	-.238	-.213	.687	.238	-.196	-.200
	.7532	.688	.170	-.256	-.209	.687	.178	-.240	-.200
	.7806	.688	.105	-.224	-.163	.712	.117	-.186	-.162
	.8628	.646	-.175	-.219	-.168	.648	-.196	-.186	-.177
	.9313	.577	-.433	-.191	-.168	.595	-.476	-.186	-.181

TABLE 29 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60^\circ$ $\delta_a = 00^\circ$ $i_t = 00^\circ$

C_{μ_f} = .100 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.4 °				α = 16.1 °			
WING									
Upper	.0000	-1.492	-6.044	-10.099	-8.326	-6.035	-15.748	-5.984	-6.765
	.0025	-2.009	-6.054	-9.887	-9.606	-5.927	-11.228	-5.931	-5.947
	.0150	-1.152				-2.626			
	.0250	-1.014	-2.550	-3.651	-3.294	-2.139	-5.026	-5.947	-6.499
	.0400	-.911	-2.302	-3.024	-2.632	-1.787	-4.131	-6.015	-6.489
	.0700	-.990	-1.969	-2.626	-2.337	-1.668	-3.157	-6.073	-6.385
	.1200	-1.339	-2.277	-2.701	-2.511	-1.937	-3.242	-5.689	-4.999
	.1900	-2.029	-2.646	-2.742	-3.569	-2.564	-3.336	-2.610	-3.807
	.2200	-1.527	-2.302	-2.398	-2.149	-2.310	-2.810	-2.557	-2.531
	.2800	-1.354	-1.974	-1.959	-1.555	-1.818	-2.326	-2.152	-2.005
	.4300	-1.118	-1.605	-1.348	-.975	-1.279	-1.805	-1.499	-1.312
	.5800	-1.330	-1.838	-.939	-.753	-1.414	-1.973	-1.063	-.989
	.6800	-1.866	-2.429	-.818	-.714	-1.875	-2.510	-.878	-.911
Lower	.0025	.049	-1.545	-1.701	-2.245	-1.580	-5.762	-2.289	-4.234
	.0050	.295	-.409	-1.262	-.767	-.575	-2.199	-1.952	-1.937
	.0100	.354	.045	-.585	-.135	.015	-1.163	-1.147	-.927
	.0250	.310	.414	.151	.246	.300	.047	-.063	-.026
	.0700	.458	.611	.323	.246	.435	.636	.278	.213
	.1200	.458	.550	.373	.275	.476	.500	.368	.244
	.2200	.450	.520	.303	.222	.490	.484	.321	.187
	.2800	.438	.404	.247	.178	.507	.442	.273	.156
	.4300	.443	.363	.055	.033	.497	.363	.089	.036
	.5800	.546	.313	-.121	-.082	.564	.315	-.078	-.083
	.7200	.625	.116	-.151	-.082	.637	.147	-.142	-.124
FLAP or AILERON									
Upper	.7312	-.050	-.062			-.054	-.067		
	.7395	-1.440	-1.411			-1.383	-1.389		
	.7532	-12.292	-19.030	.007	.002	-11.936	-19.743	-.050	-.054
	.7806	-2.615	-8.989	-.929	-.666	-2.450	-8.947	-.742	-.739
	.8354	-1.349	-3.605	-.585	-.444	-1.150	-3.547	-.589	-.619
	.9038	-.571	-2.312	-.434	-.342	-.471	-2.168	-.510	-.510
	.9440	-.472	-1.924	-.752	-.410	-.450	-1.978	-.699	-.515
	.9863	-.423	-2.045	-.505	-.338	-.398	-2.126	-.384	-.437
	.9928	-.049	-1.333	-.272	-.265	-.067	-1.378	-.331	-.380
Lower	.7312	.689	.207	-.237	-.173	.694	.226	-.173	-.171
	.7367	.694	.161	-.196	-.164	.694	.189	-.173	-.177
	.7428	.694	.186	-.196	-.154	.694	.226	-.173	-.218
	.7532	.694	.131	-.247	-.183	.694	.163	-.168	-.182
	.7806	.699	.065	-.196	-.140	.694	.100	-.168	-.156
	.8628	.655	-.282	-.196	-.159	.637	-.247	-.178	-.182
	.9313	.591	-.600	-.202	-.164	.590	-.536	-.210	-.203

TABLE 30

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60 ° δ_a = 10 ° i_t = 00 °

C_{μ_f} = .101 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.1 °				α = 8.7 °			
WING									
Upper	.0000	.327	-.697	-2.860	-2.053	-.511	-4.075	-7.853	-6.625
	.0025	.009	-1.252	-3.448	-3.179	-1.090	-4.260	-7.882	-7.887
	.0150	-.149				-.674			
	.0250	-.195	-.903	-1.666	-1.441	-.674	-1.947	-2.975	-2.931
	.0400	-.227	-.967	-1.444	-1.211	-.626	-1.819	-2.514	-2.354
	.0700	-.409	-.967	-1.328	-1.157	-.746	-1.578	-2.184	-2.116
	.1200	-.749	-1.302	-1.661	-1.531	-1.109	-1.890	-2.397	-2.378
	.1900	-1.486	-1.995	-2.059	-2.702	-1.832	-2.435	-2.567	-3.567
	.2200	-1.149	-1.688	-1.851	-1.666	-1.382	-2.018	-2.252	-2.194
	.2800	-1.045	-1.477	-1.532	-1.225	-1.253	-1.734	-1.863	-1.626
	.4300	-.945	-1.302	-1.212	-.878	-1.062	-1.445	-1.383	-1.126
	.5800	-1.199	-1.564	-.944	-.747	-1.286	-1.687	-1.058	-.970
	.6800	-1.758	-2.146	-.995	-.837	-1.860	-2.217	-1.097	-1.053
Lower	.0025	.395	.243	-.111	-.054	.325	-.829	-1.116	-1.669
	.0050	.349	.316	.074	.189	.382	-.118	-.766	-.485
	.0100	.295	.449	.203	.283	.397	.232	-.286	-.014
	.0250	.172	.449	.300	.279	.310	.464	.218	.252
	.0700	.331	.431	.296	.207	.425	.573	.344	.252
	.1200	.377	.450	.347	.243	.425	.550	.398	.247
	.2200	.380	.522	.277	.211	.420	.535	.315	.223
	.2800	.381	.238	.203	.180	.411	.500	.257	.184
	.4300	.377	.376	.060	.054	.411	.388	.097	.043
	.5800	.486	.344	-.060	-.036	.521	.341	-.004	-.038
	.7200	.613	.169	-.064	-.054	.626	.180	-.029	-.067
FLAP or AILERON									
Upper	.7312	-.682	-.476			-.762	-.092		
	.7395	-13.384	-13.180			-14.194	-13.567		
	.7532	-12.330	-17.070	-.810	-.727	-12.596	-17.859	-.204	-.175
	.7806	-2.645	-6.852	-2.069	-1.495	-2.659	-7.241	-2.198	-1.640
	.8354	-1.390	-2.550	-.916	-.702	-1.387	-2.682	-.970	-.815
	.9038	-.590	-1.573	-.597	-.405	-.574	-1.616	-.611	-.553
	.9440	-.468	-1.545	-.638	-.409	-.459	-1.616	-.684	-.533
	.9863	-.304	-1.596	-.620	-.409	-.435	-1.729	-.635	-.451
	.9928	-.072	-.981	-.287	-.270	-.066	-1.047	-.286	-.320
Lower	.7312	.672	.302	.194	.193	.693	.289	.237	.126
	.7367	.672	.266	.152	.117	.693	.265	.194	.067
	.7428	.672	.316	.120	.085	.693	.326	.174	.058
	.7532	.672	.247	.064	.036	.693	.251	.087	.014
	.7806	.686	.188	.037	.009	.693	.184	.063	.000
	.8628	.640	-.027	-.041	-.067	.660	-.075	-.014	-.097
	.9313	.572	-.307	-.092	-.085	.607	-.350	-.067	-.101

TABLE 30 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60^\circ$ $\delta_a = 10^\circ$ $i_t = 00^\circ$

$C_{\mu_f} = .101$ $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		$\alpha = 12.3°$				$\alpha = 16.1°$			
WING									
Upper	.0000	-2.666	-9.058	-14.045	-10.320	-6.569	-14.647	-6.182	-6.889
	.0025	-3.139	-8.529	-13.475	-13.330	-6.377	-11.244	-6.166	-6.255
	.0150	-1.596				-2.828			
	.0250	-1.378	-3.269	-4.363	-5.795	-2.274	-4.311	-6.187	-6.721
	.0400	-1.208	-2.867	-3.636	-5.075	-1.901	-3.632	-6.240	-6.795
	.0700	-1.218	-2.370	-3.174	-4.255	-1.756	-2.933	-6.292	-6.779
	.1200	-1.587	-2.682	-3.074	-3.125	-2.036	-3.117	-5.999	-5.779
	.1900	-2.243	-3.037	-2.940	-4.085	-2.631	-3.265	-2.722	-3.952
	.2200	-1.736	-2.555	-2.582	-2.565	-2.378	-2.693	-2.722	-2.711
	.2800	-1.502	-2.174	-2.124	-1.975	-1.927	-2.249	-2.324	-2.209
	.4300	-1.198	-1.724	-1.512	-1.365	-1.315	-1.709	-1.680	-1.539
	.5800	-1.402	-1.941	-1.184	-1.155	-1.450	-1.872	-1.298	-1.261
	.6800	-1.960	-2.518	-1.184	-1.220	-1.901	-2.377	-1.146	-1.293
Lower	.0025	-.288	-2.746	-2.601	-3.670	-1.828	-5.010	-2.402	-4.543
	.0050	.084	-.846	-1.975	-1.540	-.766	-1.790	-2.067	-2.156
	.0100	.288	-.222	-1.064	-.640	-.067	-.897	-1.251	-1.094
	.0250	.313	.391	.069	.055	.253	.168	-.083	-.073
	.0700	.452	.724	.318	.205	.393	.693	.303	.193
	.1200	.447	.650	.393	.245	.466	.600	.382	.256
	.2200	.440	.582	.318	.200	.470	.551	.335	.193
	.2800	.427	.450	.273	.180	.481	.450	.298	.162
	.4300	.447	.428	.119	.045	.481	.418	.125	.041
	.5800	.547	.380	.009	-.050	.564	.367	.005	-.062
	.7200	.636	.227	-.014	-.090	.642	.260	-.083	-.115
FLAP or AILERON									
Upper	.7312	-.082	-.106			-.084	-.107		
	.7395	-14.415	-13.985			-13.605	-13.720		
	.7532	-12.765	-18.395	-.273	-.223	-11.967	-17.976	-.210	-.189
	.7806	-2.716	-7.957	-2.587	-1.740	-2.460	-6.954	-2.099	-1.586
	.8354	-1.298	-2.873	-1.084	-.960	-1.150	-2.535	-1.020	-.968
	.9038	-.552	-1.730	-.656	-.685	-.440	-1.535	-.691	-.891
	.9440	-.487	-1.730	-.656	-.635	-.487	-1.464	-.691	-.628
	.9863	-.487	-1.878	-.626	-.530	-.569	-1.637	-.617	-.560
	.9928	-.074	-1.105	-.253	-.425	-.067	-.923	-.361	-.413
Lower	.7312	.691	.349	.293	.105	.678	.357	.162	.068
	.7367	.691	.317	.253	.040	.673	.331	.136	.010
	.7428	.686	.375	.248	.040	.673	.372	.125	.005
	.7532	.686	.301	.154	-.020	.673	.316	.041	-.057
	.7806	.686	.238	.119	-.020	.673	.250	.020	-.057
	.8628	.651	-.052	.034	-.110	.632	.010	-.083	-.146
	.9313	.592	-.375	-.039	-.155	.590	-.260	-.162	-.214

TABLE 31

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f= 60 ° δ_a= 20 ° i_t= 00 °

C_{μ_f}= .100 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α =-2.1 °				α = 8.6 °			
WING									
Upper	.0000	-.607	.022	.181	.101	-.486	-4.004	-8.714	-8.419
	.0025	.013	.373	.349	.261	-1.053	-4.192	-8.596	-9.143
	.0150	.441				-.673			
	.0250	.415	.228	.099	-.798	-.678	-1.927	-3.167	-3.294
	.0400	.375	.096	.022	-.702	-.644	-1.802	-2.655	-2.637
	.0700	.231	-.075	-.056	-.729	-.764	-1.575	-2.330	-2.362
	.1200	-.035	-.338	-.431	-1.133	-1.134	-1.903	-2.507	-2.560
	.1900	-.773	-.921	-.965	-2.454	-1.856	-2.439	-2.685	-3.767
	.2200	-.603	-.947	-.991	-1.537	-1.399	-2.024	-2.399	-2.367
	.2800	-.598	-.873	-.810	-1.142	-1.274	-1.758	-1.975	-1.787
	.4300	-.646	-.895	-.806	-.890	-1.086	-1.497	-1.542	-1.314
	.5800	-.956	-1.162	-.707	-.867	-1.322	-1.753	-1.325	-1.222
	.6800	-1.532	-1.798	-.905	-1.083	-1.880	-2.309	-1.483	-1.439
Lower	.0025	-.511	.000	.030	.239	.336	-.797	-1.365	-2.318
	.0050	-.568	.009	.056	.294	.394	-.106	-.980	-.850
	.0100	-.642	.127	.060	.261	.409	.237	-.409	-.203
	.0250	-.681	.193	.172	.243	.303	.459	.172	.203
	.0700	-.445	.281	.207	.202	.423	.580	.310	.256
	.1200	-.070	.175	.259	.243	.428	.550	.369	.270
	.2200	-.039	.272	.228	.216	.420	.531	.320	.213
	.2800	.214	.162	.198	.179	.409	.450	.276	.188
	.4300	.279	.276	.108	.064	.399	.391	.123	.077
	.5800	.445	.250	.047	.028	.519	.357	.049	-.010
	.7200	.546	.171	.052	.055	.615	.208	.079	.005
FLAP or AILERON									
Upper	.7312	-.041	-.048			-.044	-.054		
	.7395	-14.265	-12.421			-14.673	-13.591		
	.7532	-12.738	-17.233	-.309	-.285	-12.852	-18.508	-.418	-.387
	.7806	-2.589	-6.836	-2.039	-1.807	-2.644	-7.762	-2.778	-2.454
	.8354	-1.384	-2.219	-.823	-.812	-1.408	-2.497	-1.212	-1.140
	.9038	-.563	-1.022	-.478	-.495	-.586	-1.227	-.759	-.744
	.9440	-.441	-1.258	-.272	-.404	-.466	-1.594	-.645	-.613
	.9863	-.310	-1.430	-.422	-.321	-.317	-1.845	-.631	-.497
	.9928	-.070	-.890	-.168	-.239	-.062	-1.174	-.281	-.338
Lower	.7312	.642	.276	.138	.138	.678	.314	.246	.130
	.7367	.633	.250	.190	.266	.683	.270	.286	.174
	.7428	.629	.294	.190	.193	.678	.328	.281	.275
	.7532	.637	.224	.155	.225	.692	.261	.232	.121
	.7806	.655	.167	.129	.151	.697	.217	.192	.072
	.8628	.611	-.031	.052	.028	.644	-.034	.064	-.024
	.9313	.550	-.228	.004	-.037	.591	-.386	.000	-.082

TABLE 31 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f= 60 ° δ_a= 20 ° i_t= 00 °

C_{μ_f}= .100 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.3 °				α = 16.0 °			
WING									
Upper	.0000	-2.609	-8.895	-14.711	-10.925	-7.066	-16.081	-6.181	-6.984
	.0025	-3.107	-8.410	-13.820	-13.382	-6.762	-11.434	-6.165	-6.295
	.0150	-1.584				-2.907			
	.0250	-1.381	-3.171	-5.350	-6.089	-2.340	-4.875	-6.212	-6.958
	.0400	-1.223	-2.779	-4.487	-5.776	-1.928	-4.052	-6.264	-6.989
	.0700	-1.233	-2.294	-3.787	-5.234	-1.763	-3.124	-6.331	-7.015
	.1200	-1.609	-2.598	-3.208	-3.393	-2.026	-3.228	-6.129	-6.264
	.1900	-2.269	-2.975	-3.101	-4.099	-2.562	-3.378	-2.653	-4.026
	.2200	-1.761	-2.446	-2.741	-2.687	-2.304	-2.751	-2.663	-2.813
	.2800	-1.523	-2.088	-2.289	-2.119	-1.835	-2.300	-2.290	-2.331
	.4300	-1.218	-1.671	-1.706	-1.542	-1.268	-1.793	-1.679	-1.684
	.5800	-1.426	-1.902	-1.426	-1.388	-1.417	-1.938	-1.311	-1.477
	.6800	-1.964	-2.446	-1.548	-1.577	-1.866	-2.451	-1.181	-1.518
Lower	.0025	-.284	-2.691	-3.081	-3.711	-2.072	-5.839	-2.502	-4.875
	.0050	.096	-.828	-2.406	-1.687	-.866	-2.166	-2.171	-2.316
	.0100	.284	-.240	-1.320	-.751	-.088	-1.155	-1.311	-1.186
	.0250	.310	.348	-.030	.030	.263	.078	-.104	-.119
	.0700	.452	.671	.305	.239	.417	.684	.280	.192
	.1200	.447	.620	.381	.264	.500	.600	.378	.249
	.2200	.450	.559	.325	.214	.520	.528	.332	.202
	.2800	.437	.500	.289	.194	.526	.451	.306	.171
	.4300	.452	.421	.152	.070	.521	.414	.140	.036
	.5800	.558	.368	.076	-.010	.608	.373	.073	-.041
	.7200	.645	.245	.081	-.020	.685	.264	.031	-.057
FLAP or AILERON									
Upper	.7312	-.047	-.058			-.049	-.057		
	.7395	-14.447	-14.004			-13.885	-13.631		
	.7532	-12.576	-18.954	-.396	-.364	-11.957	-19.091	-.211	-.215
	.7806	-2.680	-7.739	-2.777	-2.363	-2.412	-8.062	-1.829	-1.554
	.8354	-1.284	-2.500	-1.198	-1.184	-1.124	-2.580	-1.041	-1.052
	.9038	-.528	-1.240	-.721	-.776	-.417	-1.337	-.886	-.953
	.9440	-.487	-1.603	-.635	-.627	-.459	-1.720	-.591	-.876
	.9863	-.538	-1.872	-.584	-.512	-.521	-1.974	-.803	-.901
	.9928	-.066	-1.152	-.274	-.343	-.041	-1.285	-.575	-.793
Lower	.7312	.695	.338	.284	.104	.711	.342	.150	.031
	.7367	.700	.314	.315	.119	.722	.311	.202	.078
	.7428	.700	.358	.320	.284	.737	.363	.207	.130
	.7532	.700	.289	.259	.090	.727	.295	.155	.016
	.7806	.716	.245	.208	.060	.716	.244	.104	-.005
	.8628	.655	-.034	.081	-.055	.675	-.047	-.031	-.140
	.9313	.604	-.372	-.020	-.119	.639	-.389	-.155	-.244

TABLE 32

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60°$ $\delta_a = 30°$ $i_t = 00°$

$C_{\mu_f} =$ .104 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 4.9°				α = 8.5°			
WING									
Upper	.0000	.381	-.905	-3.714	-3.550	-.750	-4.100	-10.150	-10.050
	.0025	.019	-1.453	-4.169	-4.247	-1.325	-4.474	-9.950	-10.005
	.0150	-.156				-.835			
	.0250	-.199	-.962	-1.872	-2.185	-.785	-2.093	-3.500	-3.990
	.0400	-.246	-1.033	-1.636	-1.804	-.740	-1.964	-2.911	-3.231
	.0700	-.417	-1.005	-1.481	-1.666	-.840	-1.716	-2.525	-2.837
	.1200	-.782	-1.358	-1.820	-1.995	-1.230	-2.098	-2.698	-2.931
	.1900	-1.550	-1.901	-2.231	-3.299	-1.965	-2.592	-2.856	-4.217
	.2200	-1.189	-1.745	-2.033	-2.157	-1.495	-2.263	-2.554	-2.704
	.2800	-1.104	-1.533	-1.707	-1.666	-1.350	-1.959	-2.143	-2.069
	.4300	-1.005	-1.382	-1.448	-1.314	-1.160	-1.675	-1.713	-1.581
	.5800	-1.280	-1.655	-1.335	-1.352	-1.400	-1.948	-1.535	-1.566
	.6800	-1.858	-2.264	-1.651	-1.690	-1.980	-2.598	-1.846	-1.926
Lower	.0025	.398	.179	-.264	-.771	.255	-.907	-1.708	-3.369
	.0050	.351	.288	-.061	-.143	.360	-.155	-1.262	-1.369
	.0100	.299	.439	.132	.138	.395	.227	-.599	-.493
	.0250	.161	.462	.340	.271	.305	.454	.158	.128
	.0700	.332	.453	.358	.243	.425	.598	.337	.241
	.1200	.379	.222	.406	.252	.420	.134	.411	.286
	.2200	.028	.519	.354	.229	.030	.531	.356	.217
	.2800	.379	.349	.307	.195	.410	.314	.327	.182
	.4300	.370	.410	.189	.081	.405	.407	.203	.084
	.5800	.507	.391	.151	.014	.535	.381	.153	.005
	.7200	.621	.264	.189	.048	.630	.263	.173	.010
FLAP or AILERON									
Upper	.7312	-.810	-.816			-.900	-1.062		
	.7395	-14.923	-13.365	-2.264	-1.947	-15.325	-14.802	-2.297	-2.098
	.7532	-13.004	-18.704	-6.230	-6.061	-13.305	-19.688	-6.400	-6.029
	.7806	-2.744	-7.970	-3.438	-3.299	-2.765	-8.009	-3.406	-3.310
	.8354	-1.502	-2.353	-1.363	-1.414	-1.485	-2.464	-1.396	-1.557
	.9038	-.630	-.858	-.745	-.843	-.635	-.907	-.792	-.956
	.9440	-.479	-1.490	-.632	-.686	-.500	-1.443	-.673	-.734
	.9863	-.441	-1.731	-.604	-.590	-.610	-1.737	-.549	-.586
	.9928	-.090	-1.250	-.245	-.362	-.075	-1.154	-.223	-.384
Lower	.7312	.682	.335	.241	.086	.695	.397	.252	.054
	.7367	.697	.311	.283	.143	.705	.350	.312	.089
	.7428	.697	.311	.321	.262	.700	.402	.351	.276
	.7532	.692	.288	.321	.167	.695	.340	.332	.103
	.7806	.701	.241	.278	.129	.720	.283	.297	.084
	.8628	.668	.038	.156	.024	.665	.098	.198	-.039
	.9313	.602	-.269	.075	-.048	.615	-.247	.109	-.094

TABLE 32 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

$\delta_f = 60^{\circ}$ $\delta_a = 30^{\circ}$ $i_t = 00^{\circ}$

C_{μ_f} = .104 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.3 °				α = 16.0 °			
WING									
Upper	.0000	-3.053	-9.421	-15.350	-8.789	-4.360	-13.320	-6.300	-6.950
	.0025	-3.397	-8.900	-14.893	-12.877	-4.616	-10.647	-6.461	-6.294
	.0150	-1.714				-2.197			
	.0250	-1.482	-3.344	-5.643	-6.444	-1.844	-4.093	-6.274	-6.477
	.0400	-1.296	-2.937	-4.872	-6.316	-1.585	-3.466	-6.321	-6.538
	.0700	-1.296	-2.417	-4.133	-5.913	-1.502	-2.813	-6.290	-6.477
	.1200	-1.663	-2.734	-3.342	-3.913	-1.844	-3.067	-5.077	-5.112
	.1900	-2.327	-3.073	-3.168	-4.219	-2.487	-3.248	-2.803	-4.035
	.2200	-1.829	-2.562	-2.842	-2.760	-2.067	-2.694	-2.751	-2.726
	.2800	-1.568	-2.198	-2.388	-2.194	-1.658	-2.274	-2.331	-2.198
	.4300	-1.271	-1.786	-1.775	-1.617	-1.290	-1.777	-1.730	-1.604
	.5800	-1.462	-2.021	-1.469	-1.393	-1.487	-1.984	-1.409	-1.396
	.6800	-2.025	-2.620	-1.541	-1.531	-2.015	-2.513	-1.394	-1.492
Lower	.0025	-.407	-2.932	-3.270	-4.020	-.927	-4.596	-2.311	-4.315
	.0050	-.005	-.937	-2.571	-1.801	-.264	-1.679	-1.948	-2.005
	.0100	.241	-.312	-1.393	-.832	.145	-.819	-1.119	-.959
	.0250	.302	.344	-.046	.026	.306	.155	-.026	-.030
	.0700	.457	.682	.316	.235	.461	.674	.306	.218
	.1200	.417	.042	.403	.270	.456	.570	.414	.259
	.2200	.015	.552	.347	.209	.021	.523	.378	.193
	.2800	.432	.385	.311	.189	.466	.383	.342	.178
	.4300	.442	.427	.173	.061	.471	.414	.187	.071
	.5800	.553	.380	.092	-.031	.570	.383	.109	-.025
	.7200	.638	.281	.087	-.015	.648	.275	.078	.000
FLAP or AILERON									
Upper	.7312	-.950	-1.057			-.969	-1.119		
	.7395	-14.980	-14.827	-1.883	-1.306	-14.849	-14.056	-1.627	-1.279
	.7532	-12.959	-20.108	-3.643	-2.985	-12.797	-19.382	-2.938	-2.477
	.7806	-2.754	-8.343	-2.413	-1.393	-2.684	-8.031	-2.145	-1.320
	.8354	-1.342	-2.552	-1.372	-1.224	-1.238	-2.456	-1.321	-1.173
	.9038	-.623	-1.031	-1.138	-1.291	-.596	-1.311	-1.192	-1.279
	.9440	-.518	-1.656	-.612	-1.316	-.497	-1.730	-.611	-1.269
	.9863	-.568	-1.984	-1.189	-1.255	-1.233	-1.922	-1.233	-1.284
	.9928	-.116	-1.364	-.929	-1.128	-.088	-1.399	-.995	-1.147
Lower	.7312	.668	.385	.143	-.005	.684	.357	.124	.000
	.7367	.678	.344	.194	.026	.689	.337	.166	.051
	.7428	.688	.406	.260	.235	.699	.383	.212	.168
	.7532	.688	.328	.255	.066	.699	.326	.212	.086
	.7806	.698	.281	.184	.036	.699	.275	.155	.051
	.8628	.648	.026	.020	-.107	.648	.016	-.005	-.112
	.9313	.593	-.354	-.163	-.255	.606	-.321	-.176	-.269

TABLE 33

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f = 60° δ_a = 40° i_t = 00°

C_{μ_f} = .101 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 5.0°				α = 8.6°			
WING									
Upper	.0000	.428	-.882	-3.502	-3.126	-.695	-3.894	-8.389	-8.435
	.0025	.166	-1.321	-3.905	-3.798	-1.179	-4.047	-8.274	-8.484
	.0150	-.048				-.700			
	.0250	-.096	-.856	-1.710	-1.825	-.646	-1.780	-2.898	-3.391
	.0400	-.127	-.894	-1.463	-1.493	-.596	-1.653	-2.399	-2.715
	.0700	-.297	-.869	-1.307	-1.397	-.695	-1.434	-2.061	-2.373
	.1200	-.633	-1.164	-1.632	-1.685	-1.049	-1.697	-2.219	-2.449
	.1900	-1.345	-1.612	-1.996	-2.895	-1.695	-2.118	-2.350	-3.524
	.2200	-1.013	-1.536	-1.792	-1.812	-1.291	-1.833	-2.096	-2.178
	.2800	-.934	-1.316	-1.494	-1.353	-1.143	-1.587	-1.719	-1.635
	.4300	-.838	-1.156	-1.203	-.974	-.986	-1.324	-1.294	-1.142
	.5800	-1.100	-1.380	-1.009	-.908	-1.197	-1.526	-1.074	-1.044
	.6800	-1.620	-1.915	-1.147	-1.070	-1.722	-2.052	-1.175	-1.200
Lower	.0025	.406	.232	-.208	-.572	.300	-.785	-1.263	-2.795
	.0050	.362	.321	-.004	.009	.395	-.075	-.886	-1.084
	.0100	.323	.451	.190	.240	.417	.254	-.390	-.342
	.0250	.188	.481	.364	.332	.345	.460	.281	.191
	.0700	.358	.477	.394	.288	.471	.601	.417	.302
	.1200	.410	.500	.442	.288	.471	.580	.478	.302
	.2200	.410	.532	.394	.275	.460	.561	.434	.271
	.2800	.410	.480	.359	.244	.448	.500	.390	.258
	.4300	.410	.430	.242	.135	.448	.439	.272	.151
	.5800	.528	.426	.203	.074	.547	.425	.246	.080
	.7200	.616	.312	.212	.122	.646	.320	.241	.102
FLAP or AILERON									
Upper	.7312	-.037	-.040			-.042	-.044		
	.7395	-13.412	-11.910	-.181	-.126	-13.456	-12.158	-.182	-.142
	.7532	-11.757	-16.512	-.229	-.226	-11.758	-17.002	-.222	-.218
	.7806	-2.436	-6.556	-1.442	-1.061	-2.426	-7.095	-1.416	-1.164
	.8354	-1.305	-1.949	-1.017	-1.004	-1.273	-2.070	-1.026	-1.067
	.9038	-.498	-1.013	-1.022	-1.166	-.484	-1.136	-1.044	-1.235
	.9440	-.375	-1.156	-.900	-1.135	-.386	-1.302	-.900	-1.093
	.9863	-.790	-1.283	-1.004	-1.179	-1.170	-1.399	-1.026	-1.191
	.9928	.000	-1.126	-.965	-1.078	.000	-1.289	-.978	-1.147
Lower	.7312	.690	.413	.238	.114	.686	.395	.267	.124
	.7367	.690	.384	.238	.135	.686	.373	.276	.120
	.7428	.694	.405	.273	.240	.686	.373	.307	.236
	.7532	.694	.367	.320	.196	.686	.342	.333	.151
	.7806	.694	.329	.277	.175	.704	.320	.298	.151
	.8628	.664	.207	.130	.031	.664	.184	.167	.027
	.9313	.607	-.046	-.030	-.127	.614	-.088	.000	-.142

TABLE 33 concluded

PRESSURE COEFFICIENTS FOR WING, FLAP OR AILERON

Wing leading edge configuration: 40° Droop

δ_f= 60 ° δ_a= 40 ° i_t= 00 °

C_{μ_f}= .101 $C_{\mu,LE}$: I = .000 ,C = .000 ,O = .000 $C_{\mu,K}$: I = .000 ,C = .008 ,O = .014

Surface	x/c	c_p for values of $\frac{y}{b/2}$ of -							
		0.31	0.56	0.76	0.90	0.31	0.56	0.76	0.90
		α = 12.3 °				α = 16.0 °			
WING									
Upper	.0000	-2.806	-8.860	-14.413	-7.563	-6.112	-14.757	-5.508	-5.698
	.0025	-3.207	-8.153	-13.469	-12.051	-5.950	-10.091	-5.504	-5.476
	.0150	-1.562				-2.563			
	.0250	-1.322	-3.009	-5.189	-5.711	-2.058	-4.680	-5.499	-5.346
	.0400	-1.134	-2.619	-4.430	-5.637	-1.730	-3.837	-5.574	-5.319
	.0700	-1.138	-2.153	-3.708	-5.358	-1.578	-2.914	-5.620	-5.291
	.1200	-1.465	-2.409	-2.893	-3.591	-1.828	-2.966	-5.448	-5.129
	.1900	-2.069	-2.646	-2.764	-3.707	-2.372	-3.023	-2.275	-2.805
	.2200	-1.599	-2.260	-2.467	-2.437	-2.152	-2.533	-2.294	-2.051
	.2800	-1.355	-1.916	-2.060	-1.926	-1.701	-2.071	-1.986	-1.662
	.4300	-1.064	-1.516	-1.472	-1.423	-1.171	-1.595	-1.472	-1.171
	.5800	-1.240	-1.707	-1.213	-1.228	-1.294	-1.709	-1.196	-1.273
	.6800	-1.746	-2.228	-1.245	-1.302	-1.720	-2.171	-1.149	-1.273
Lower	.0025	-.336	-2.730	-2.912	-3.581	-1.657	-5.504	-2.219	-3.902
	.0050	.074	-.856	-2.264	-1.577	-.647	-2.000	-1.902	-1.856
	.0100	.300	-.265	-1.213	-.702	.000	-1.062	-1.121	-.921
	.0250	.364	.344	.051	.084	.323	.114	-.014	.005
	.0700	.502	.679	.393	.284	.441	.695	.369	.273
	.1200	.502	.620	.463	.298	.495	.630	.458	.338
	.2200	.500	.572	.440	.260	.500	.571	.425	.301
	.2800	.502	.500	.417	.247	.515	.486	.402	.268
	.4300	.493	.460	.296	.130	.515	.486	.266	.190
	.5800	.594	.446	.241	.065	.598	.462	.215	.074
	.7200	.645	.358	.204	.107	.676	.381	.187	.102
FLAP or AILERON									
Upper	.7312	-.046	-.053			-.050	-.055		
	.7395	-13.529	-12.455	-.182	-.131	-12.192	-11.604	-.166	-.153
	.7532	-11.655	-17.390	-.197	-.178	-10.429	-16.479	-.186	-.168
	.7806	-2.415	-7.358	-1.620	-1.046	-2.245	-7.118	-1.640	-1.259
	.8354	-1.111	-2.126	-1.069	-1.037	-1.024	-1.938	-1.028	-1.074
	.9038	-.484	-1.191	-1.092	-1.135	-.368	-1.157	-1.037	-1.032
	.9440	-.382	-1.307	-.900	-1.130	-.368	-1.190	-.900	-.829
	.9863	-1.217	-1.470	-1.074	-1.130	-1.196	-1.357	-1.014	-1.000
	.9928	-.005	-1.293	-1.074	-1.149	-.010	-1.181	-.916	-.949
Lower	.7312	.705	.428	.227	.107	.696	.433	.234	.106
	.7367	.705	.391	.255	.107	.696	.433	.229	.134
	.7428	.703	.442	.287	.256	.696	.462	.229	.213
	.7532	.700	.405	.287	.153	.696	.424	.229	.171
	.7806	.700	.367	.268	.140	.696	.381	.220	.153
	.8628	.673	.195	.116	.023	.657	.200	.098	.023
	.9313	.613	-.098	-.093	-.140	.622	-.071	-.047	-.153

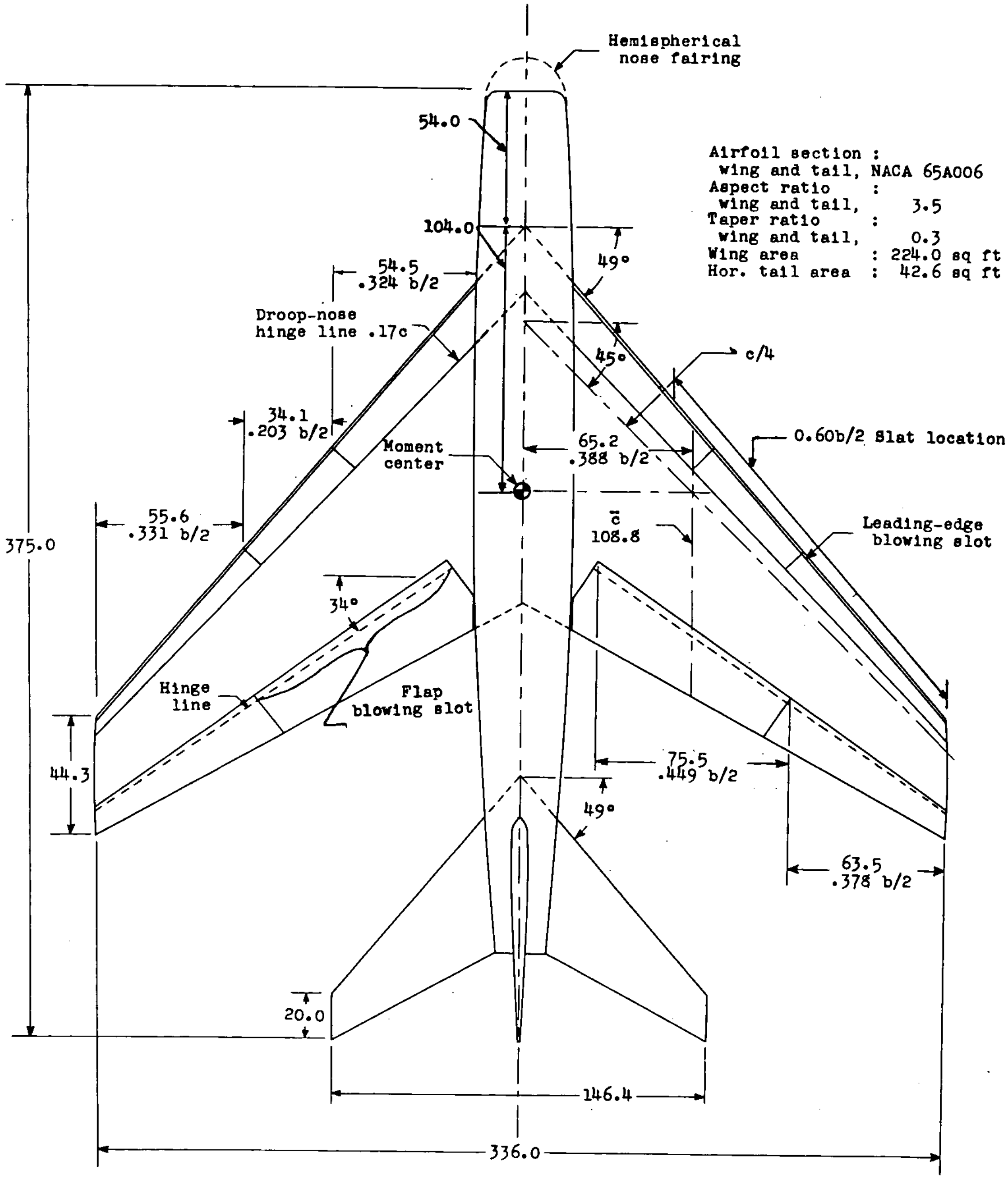
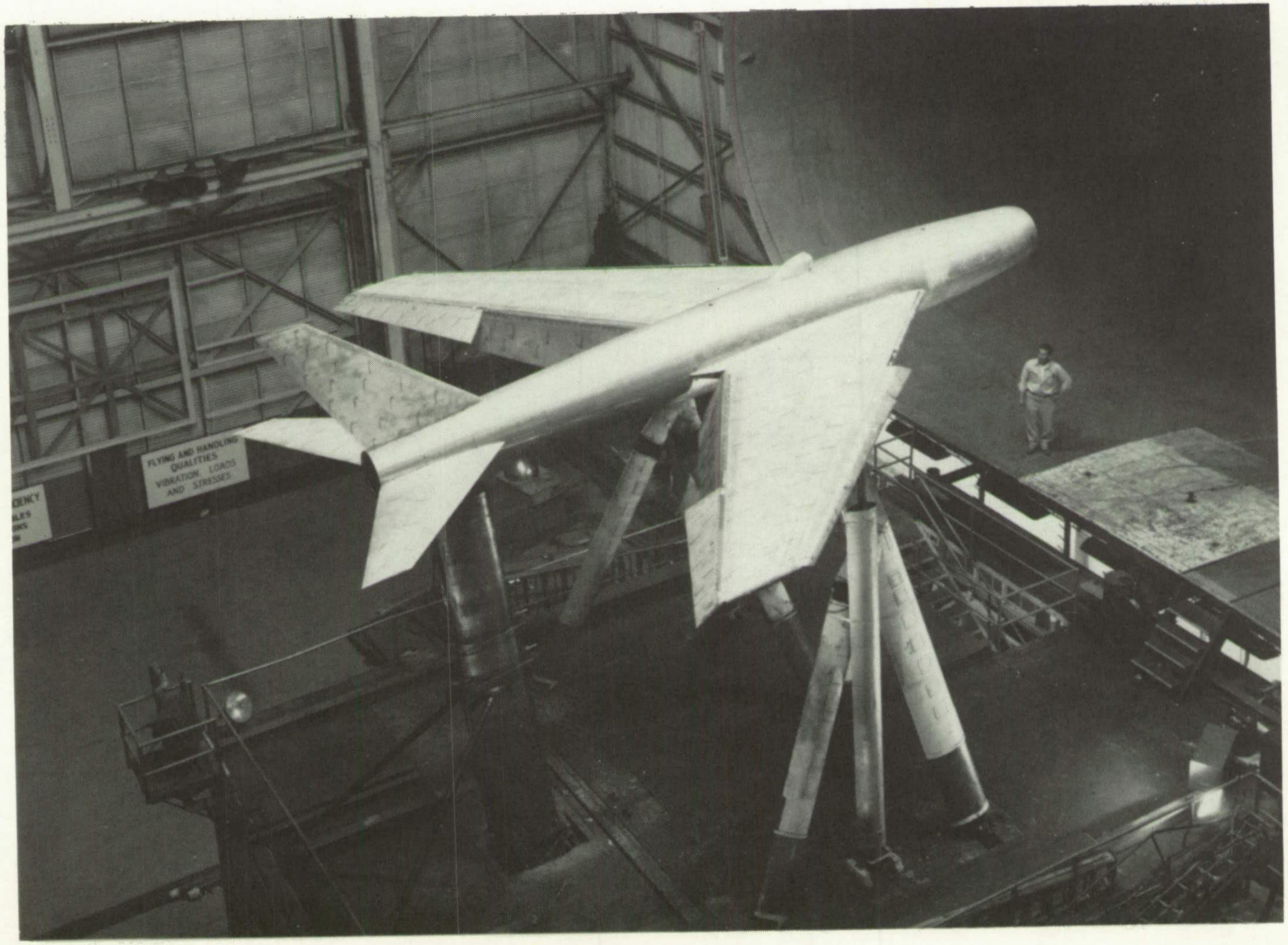


Figure 1.- Geometric characteristics of the model. All dimensions are in inches unless otherwise indicated.



L-88820
Figure 2.- General view of model mounted for tests in the Langley full-scale tunnel with flow-survey tufts attached.

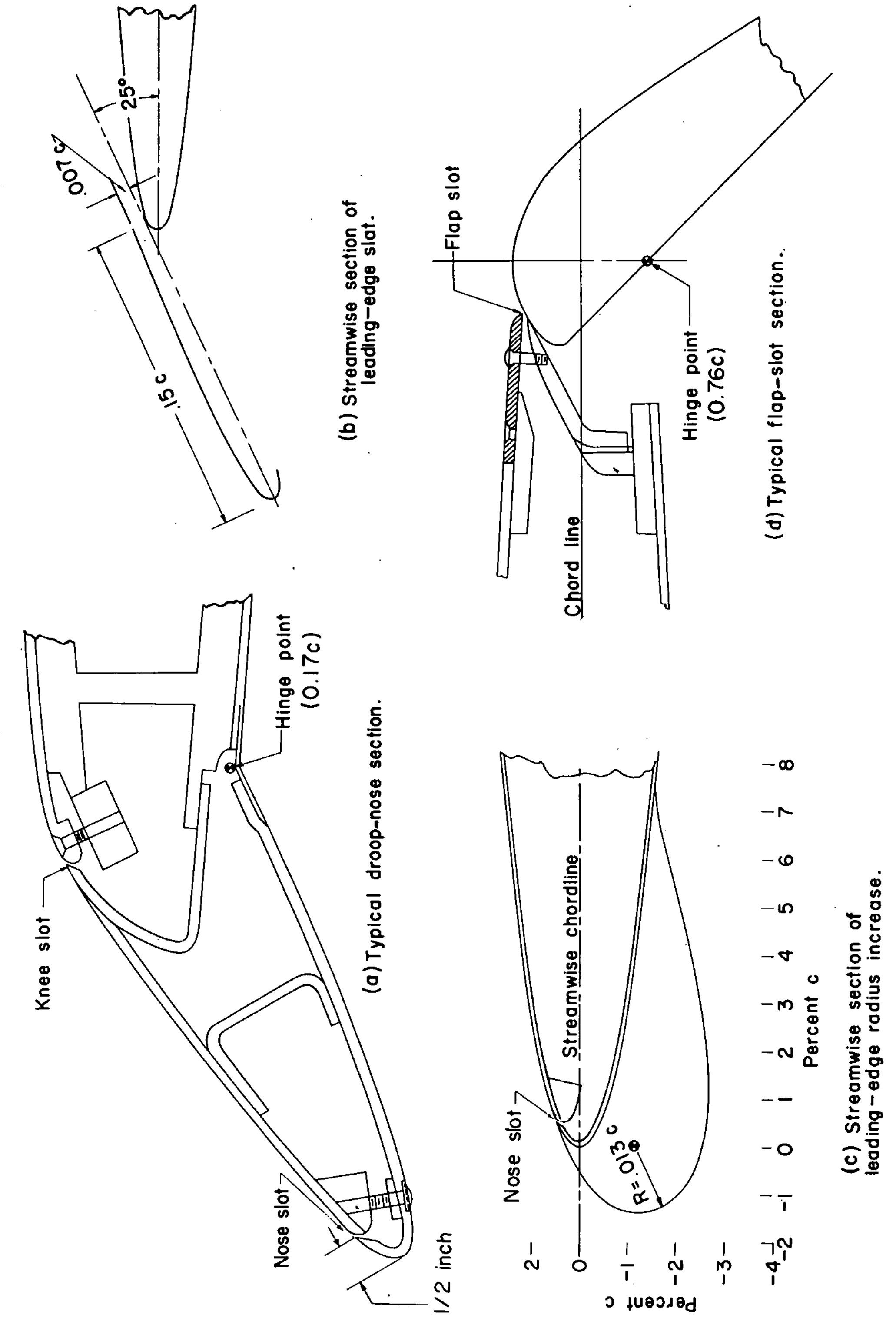


Figure 3.- Sectional views of the high-lift and flow-control devices.

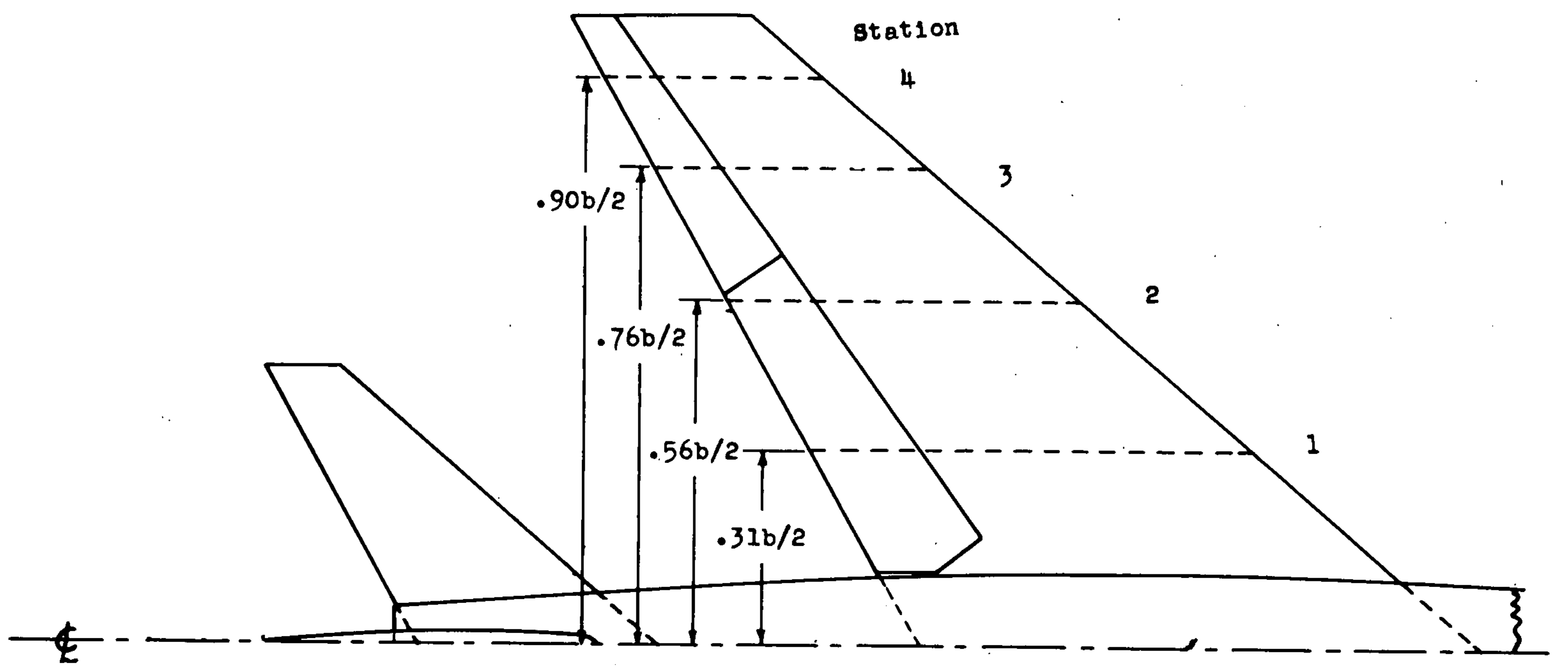


Figure 4.- Spanwise locations of pressure orifices on wing. Chordwise locations are listed in appropriate data tables.

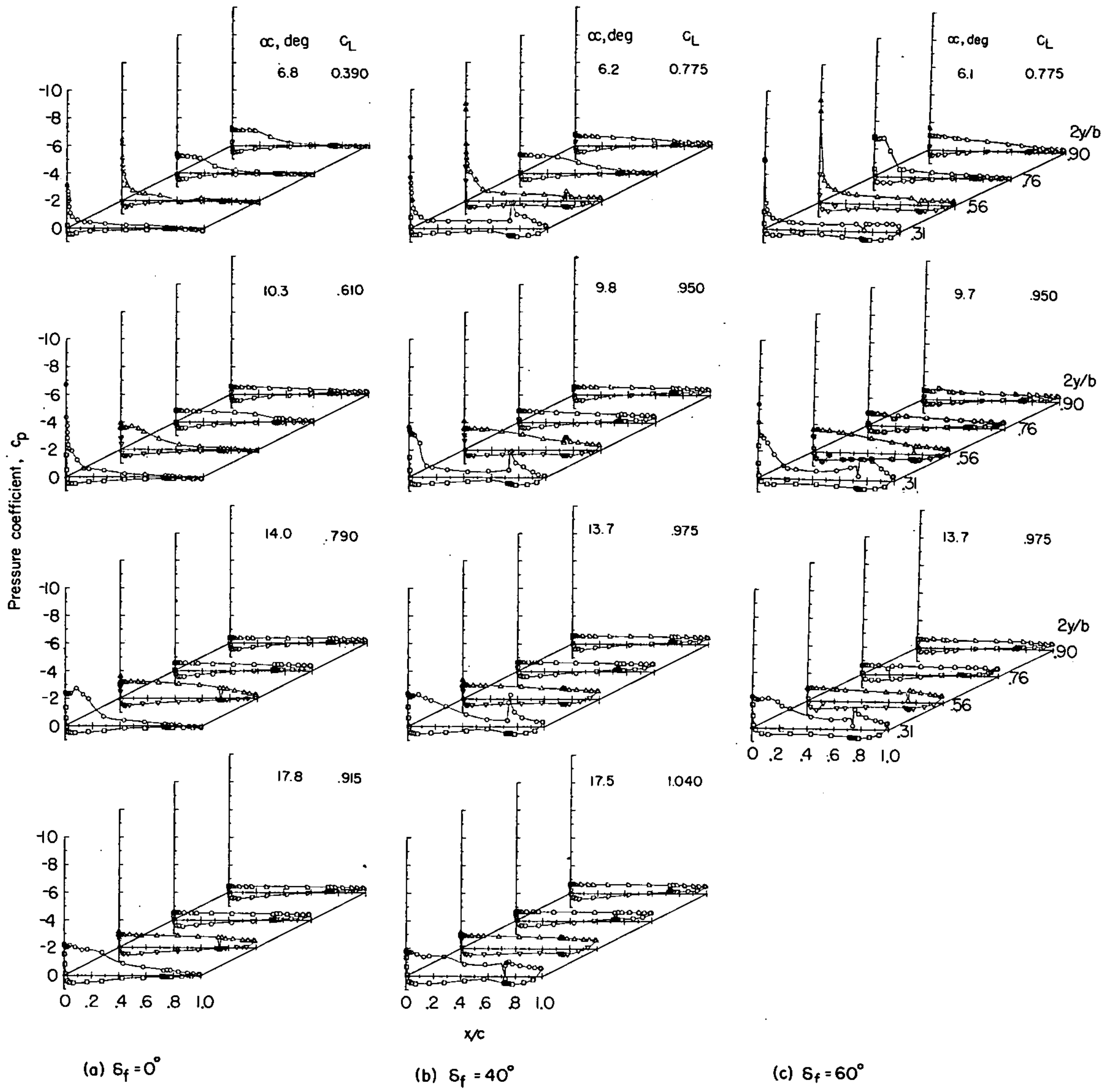


Figure 5.- Effect of flap deflection on chordwise pressure distributions for the basic wing-leading-edge configuration. $C_{\mu,f} = 0$; $\delta_a = 0°$.

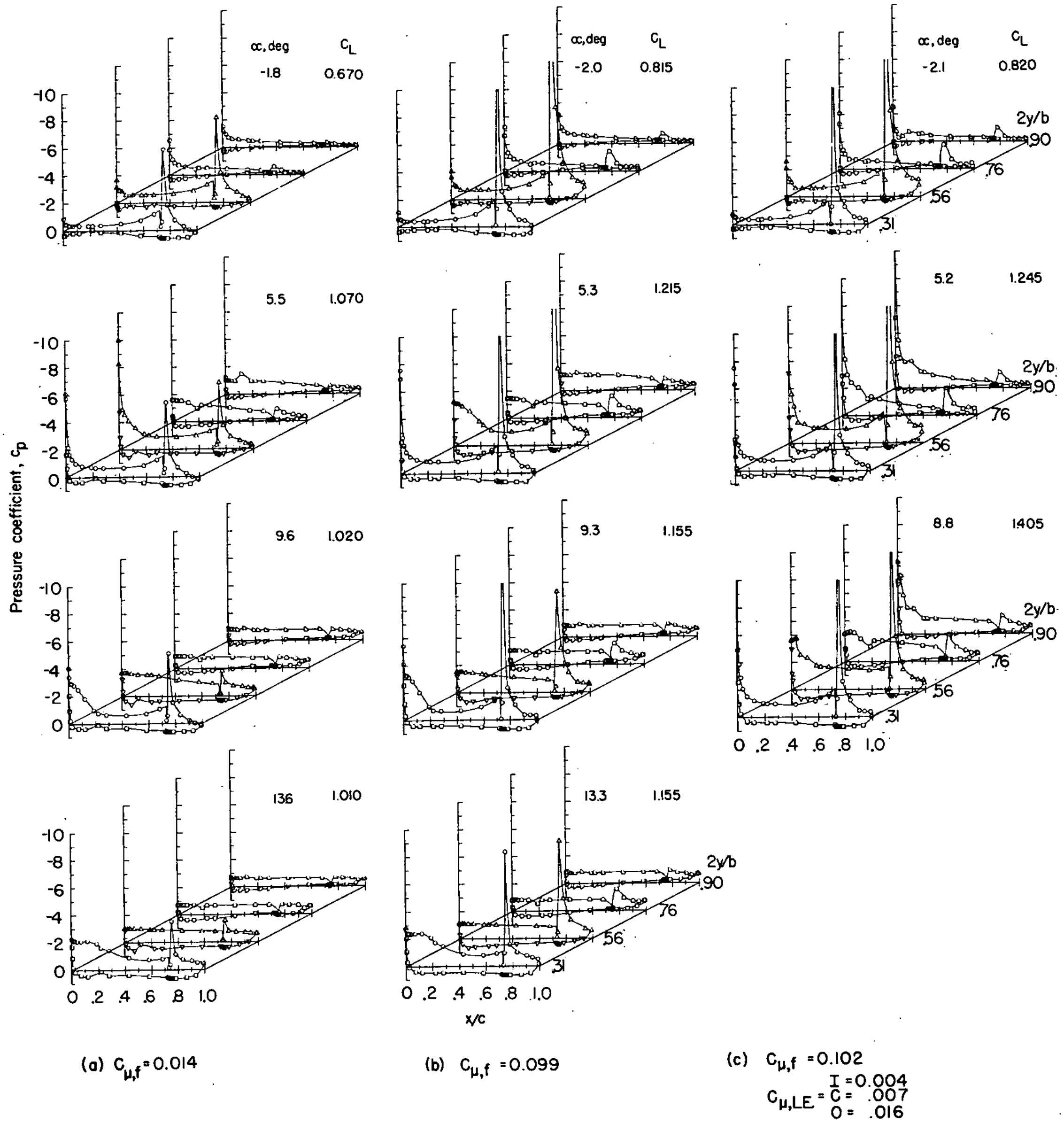
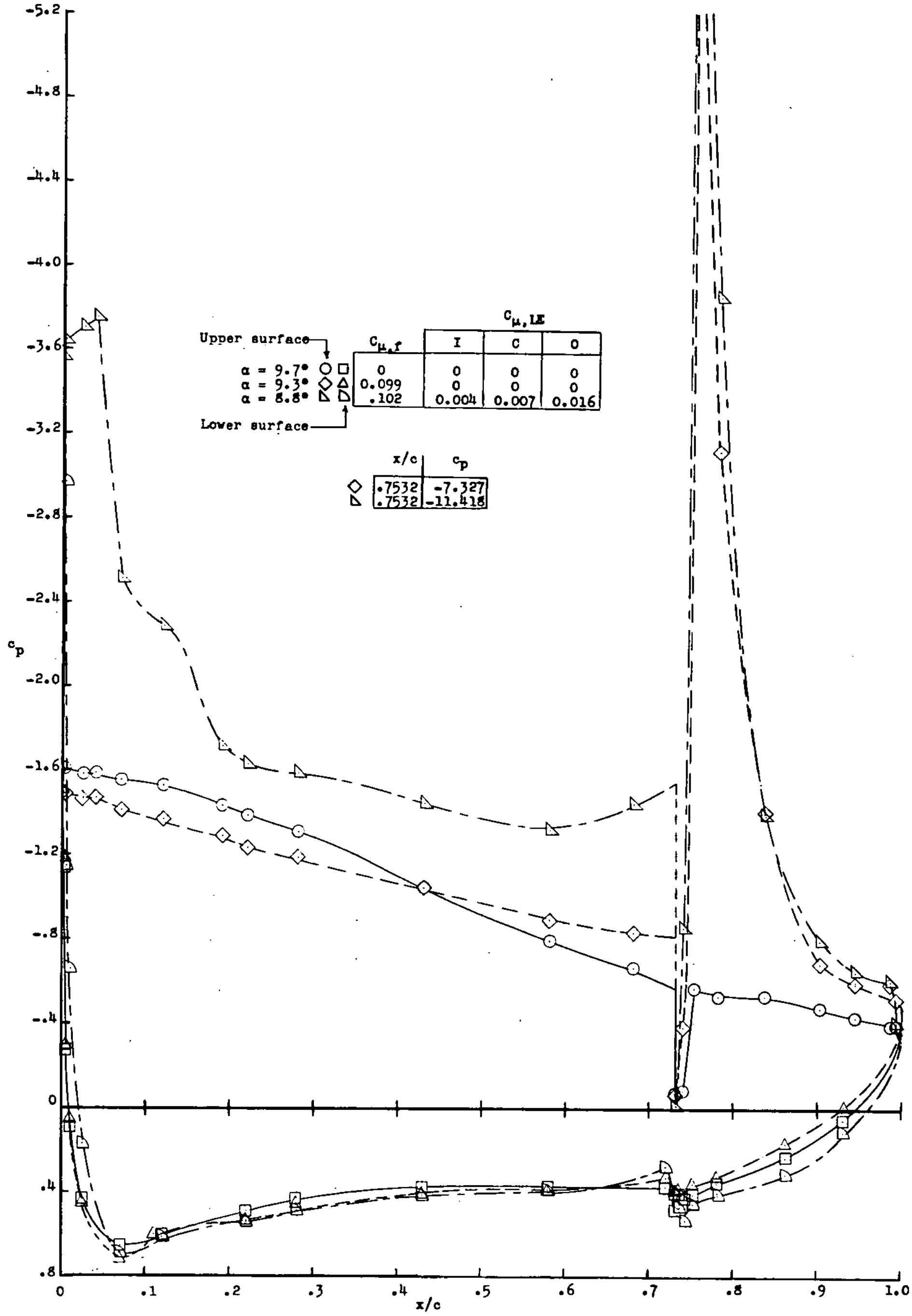
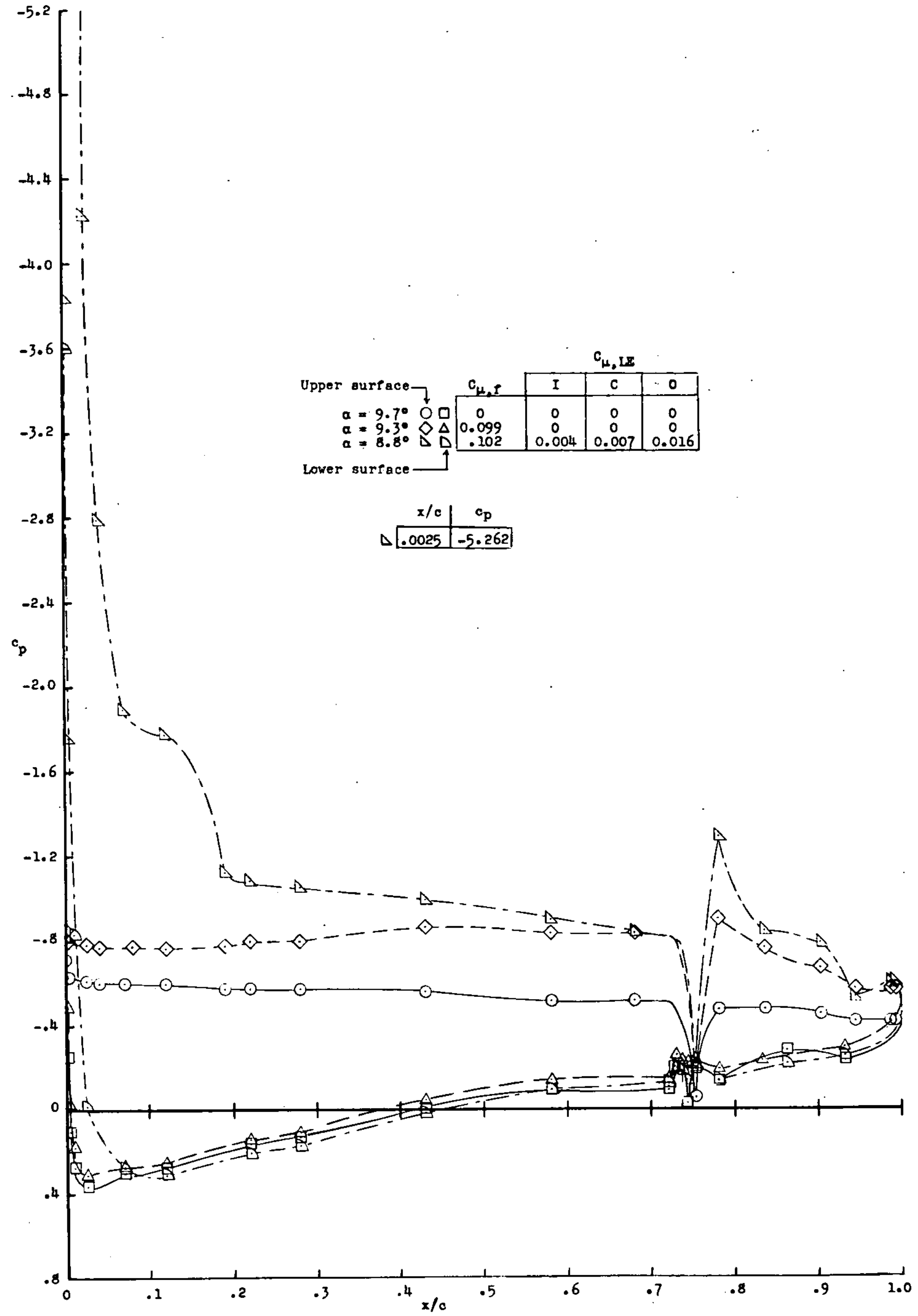


Figure 6.- Effect of flap blowing alone and in combination with leading-edge blowing on chordwise pressure distributions for the basic wing-leading-edge configuration. $\delta_f = 60^\circ$; $\delta_a = 0^\circ$.



(a) Station 2.

Figure 7.- Comparison of chordwise pressure distributions at stations 2 and 4 without and with leading-edge and flap blowing applied for the basic wing-leading-edge configuration. $\delta_f = 60^\circ$; $\delta_a = 0^\circ$.



(b) Station 4.

Figure 7.- Concluded.

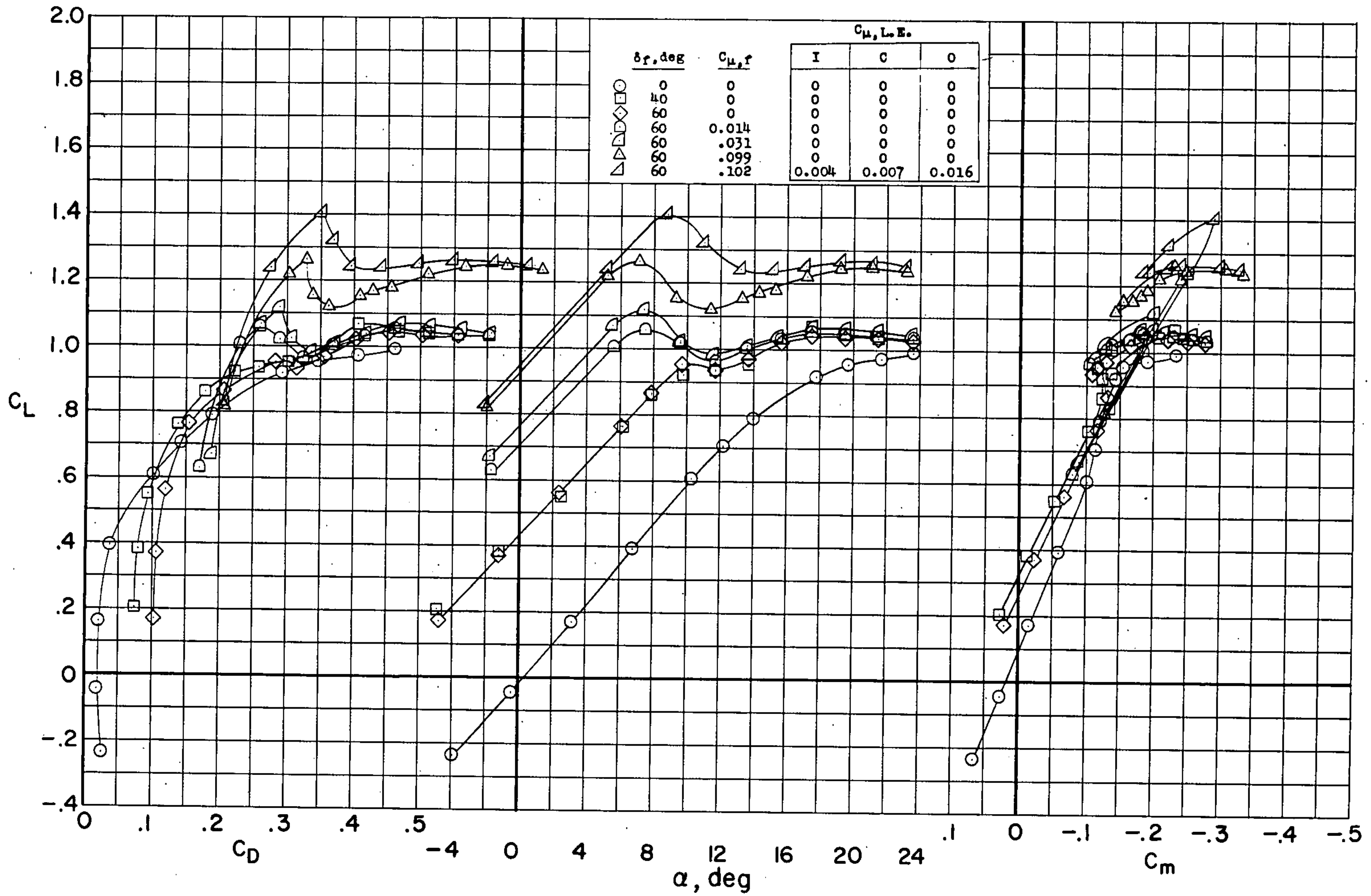


Figure 8.- Effect of flap deflection without flap blowing applied and effect of flap blowing alone and in combination with leading-edge blowing on the aerodynamic characteristics of the basic wing configuration. $\delta_a = 0^{\circ}$.

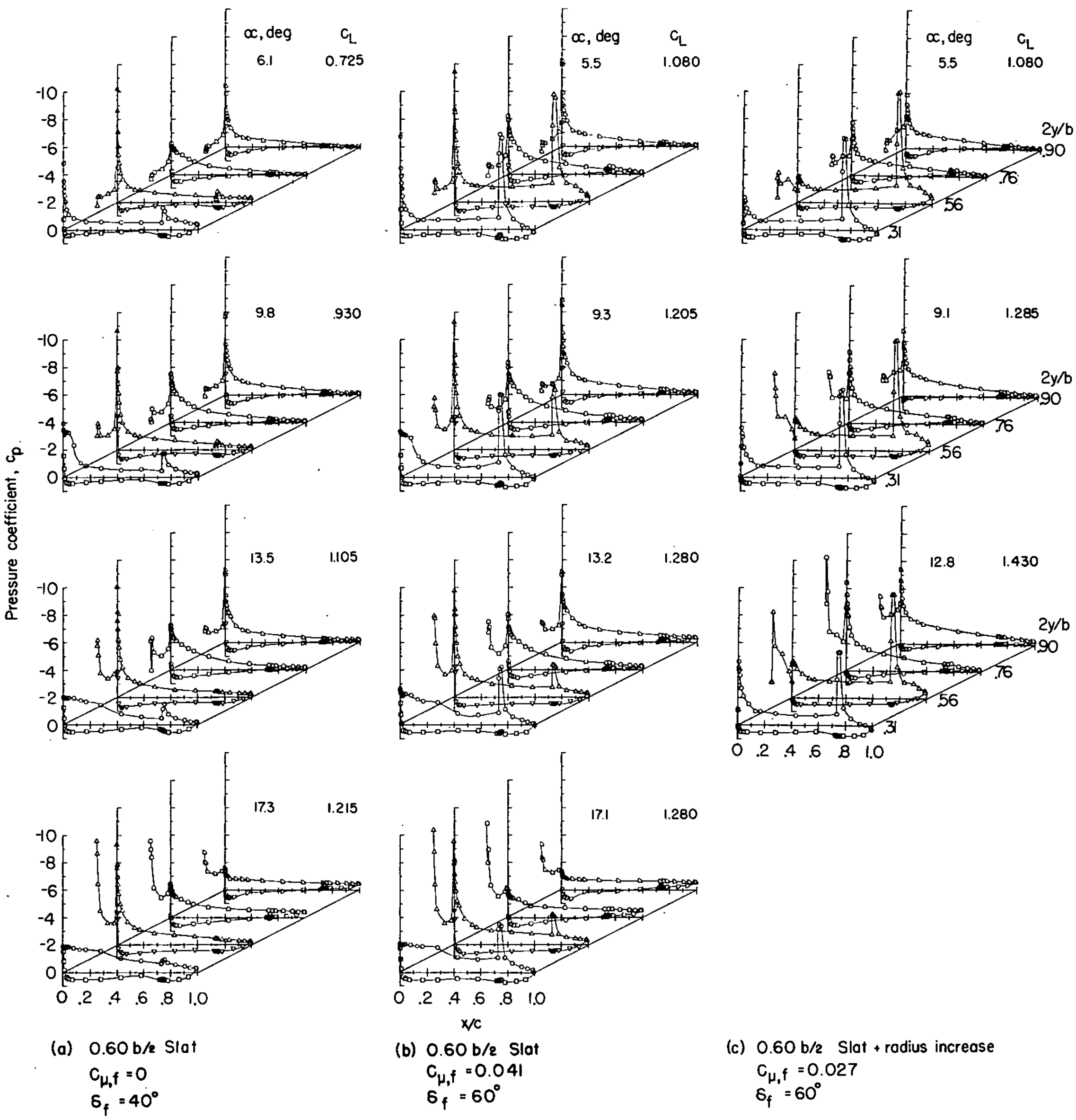
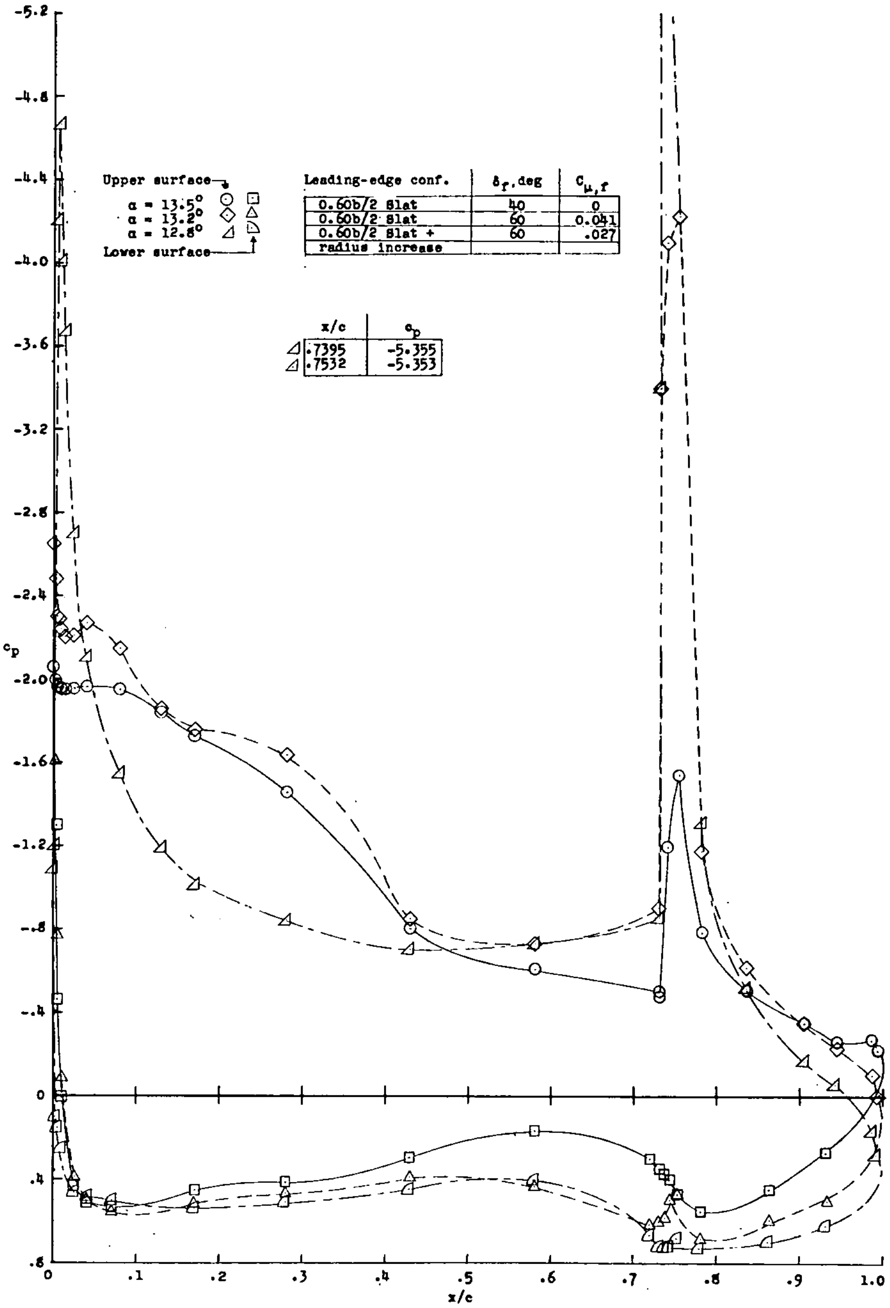
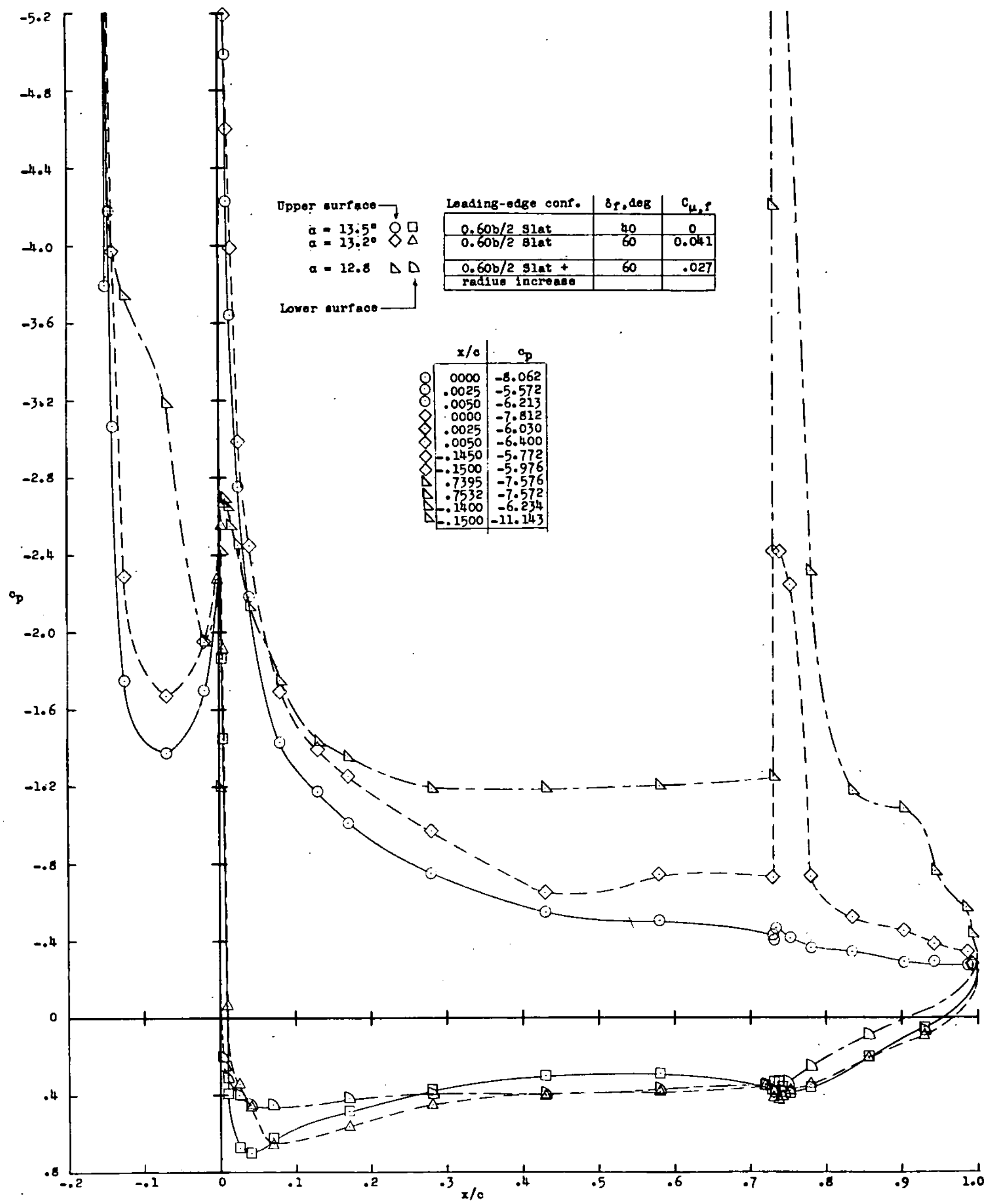


Figure 9.- Effect of flap blowing and increase of wing-leading-edge radius on chordwise pressure distributions for the 0.60b/2-slat configuration. $\delta_a = 0^\circ$.



(a) Station 1.

Figure 10.- Comparison of chordwise pressure distributions at stations 1 and 2 with the 0.60b/2 slat installed, without and with flap blowing applied, and with the 0.60b/2 slat plus radius increase installed with flap blowing applied. $\delta_a = 0^{\circ}$.



(b) Station 2.

Figure 10.- Concluded.

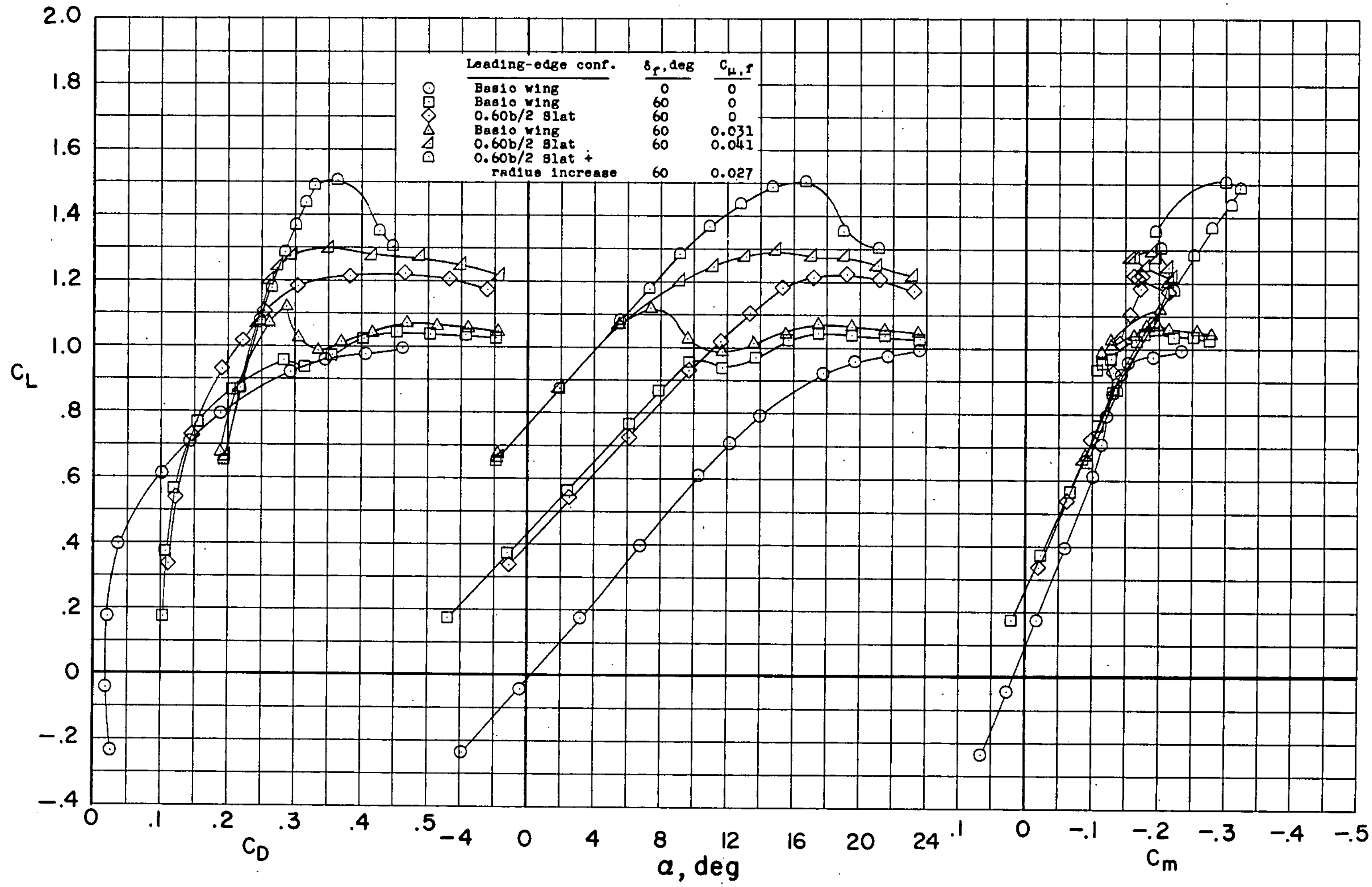


Figure 11.- Effect on aerodynamic characteristics of installing a 0.60b/2 slat without and with flap blowing applied and effect of increasing the leading-edge radius from the slat to the to the fuselage juncture with flap blowing applied. $\delta_a = 0^{\circ}$.

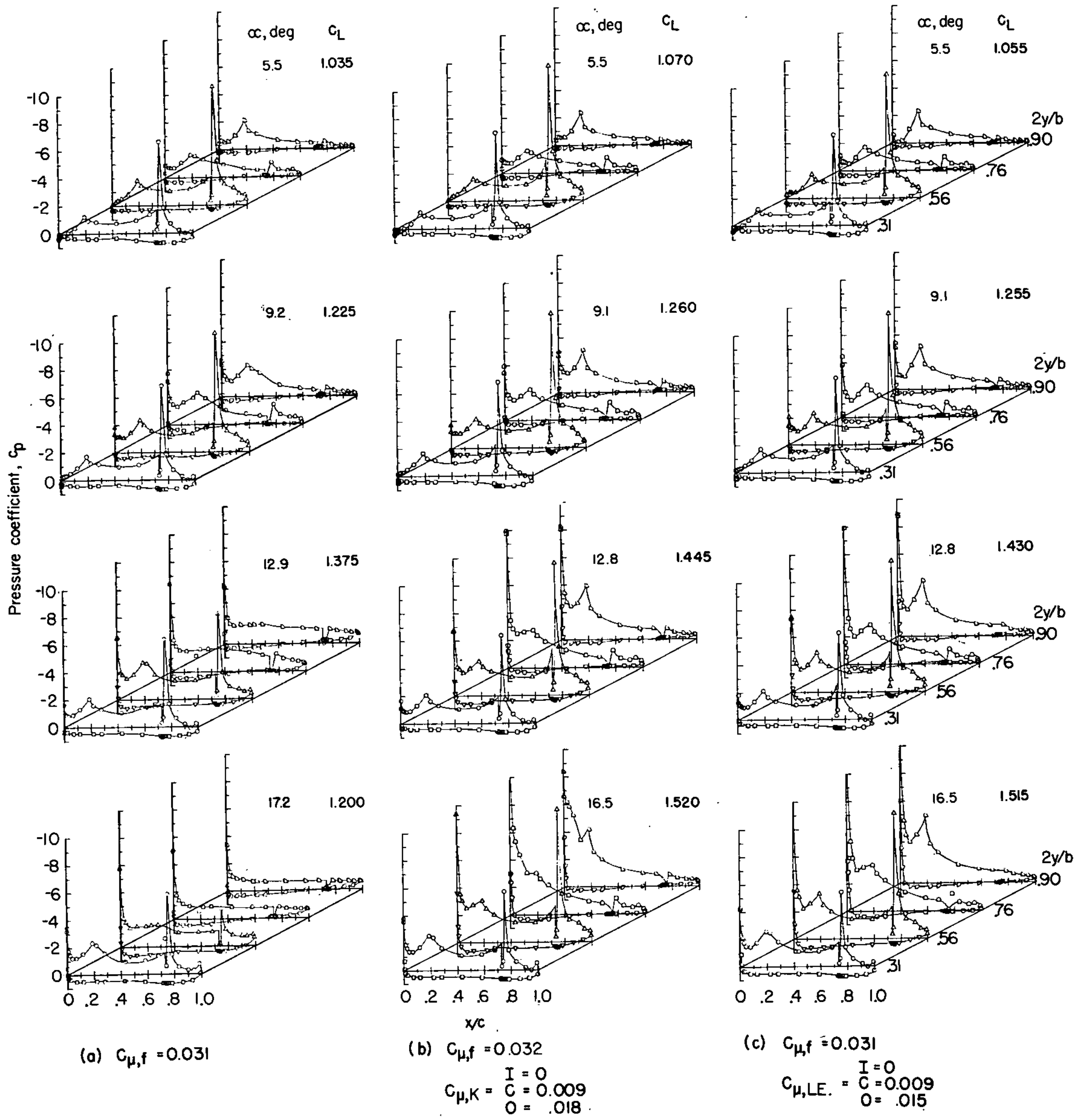
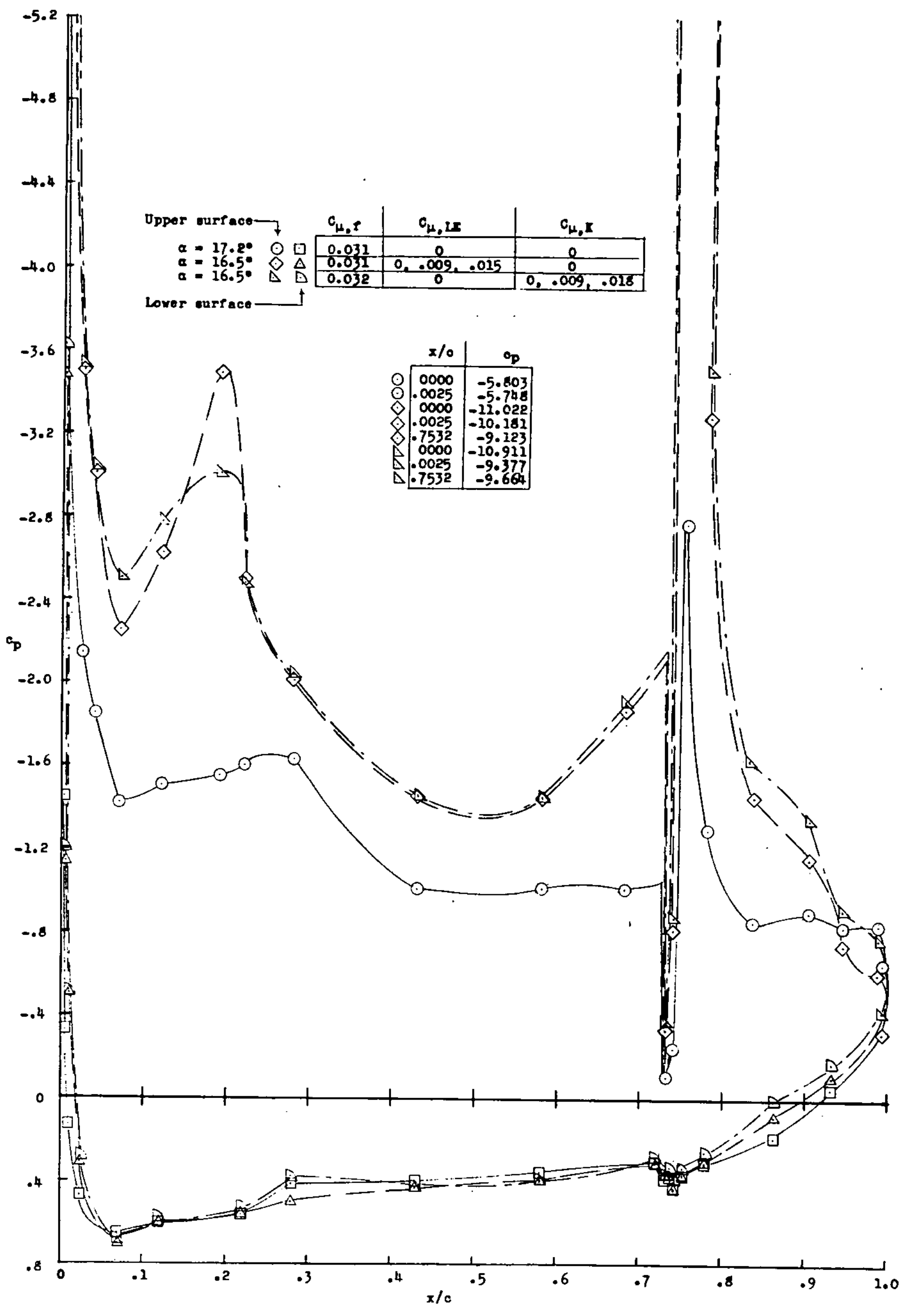
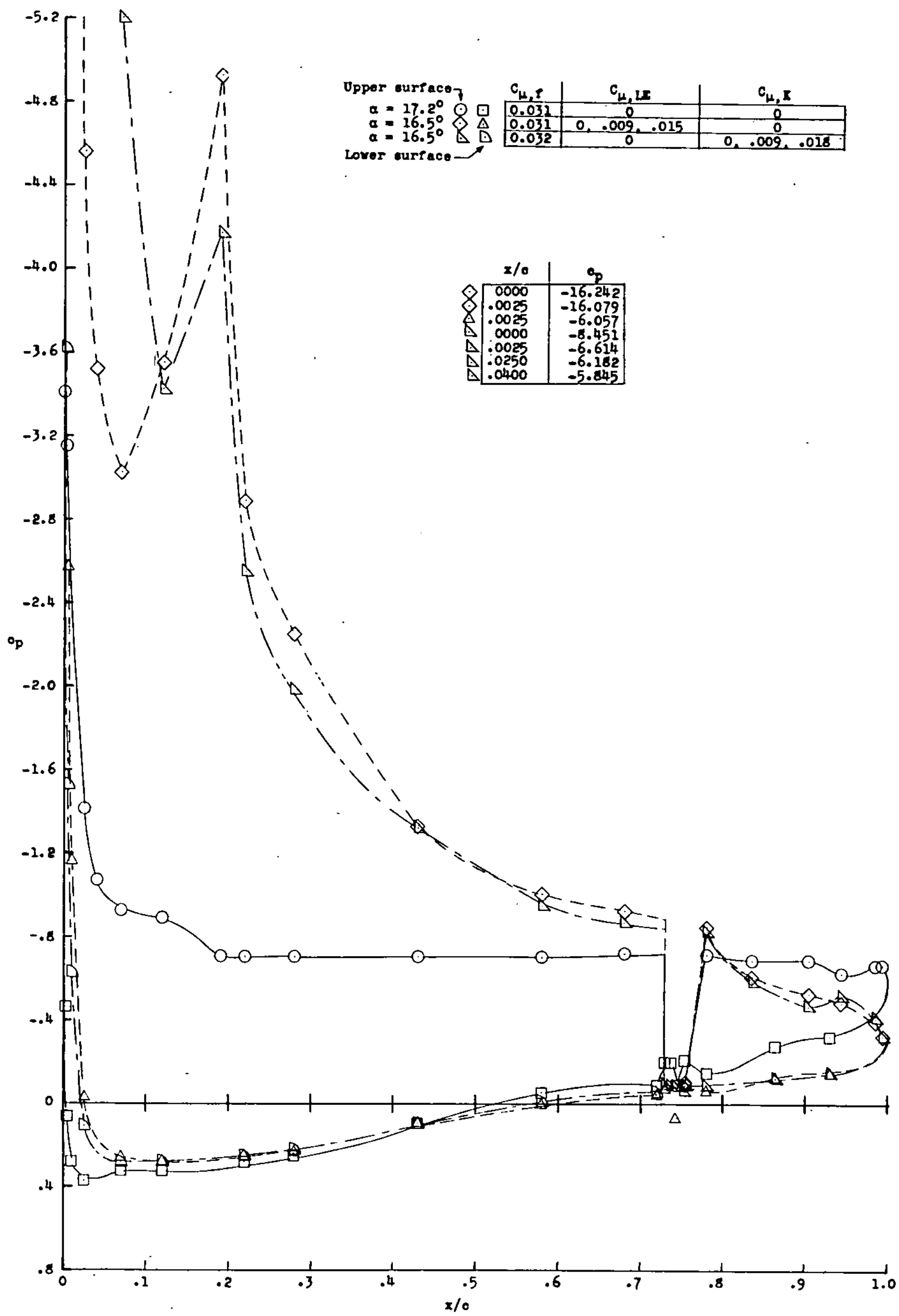


Figure 12.- Effect of partial-span leading-edge or knee blowing in combination with flap blowing on chordwise pressure distributions. Wing leading edge drooped 45°; $\delta_f = 60^\circ$; $\delta_a = 0^\circ$.



(a) Station 2.

Figure 13.- Variation of chordwise pressure distributions at stations 2 and 4 without and with leading-edge or knee blowing applied in combination with flap blowing for the 45° drooped-leading-edge configuration. $\delta_f = 60^{\circ}$; $\delta_a = 0^{\circ}$.



(b) Station 4.

Figure 13.- Concluded.

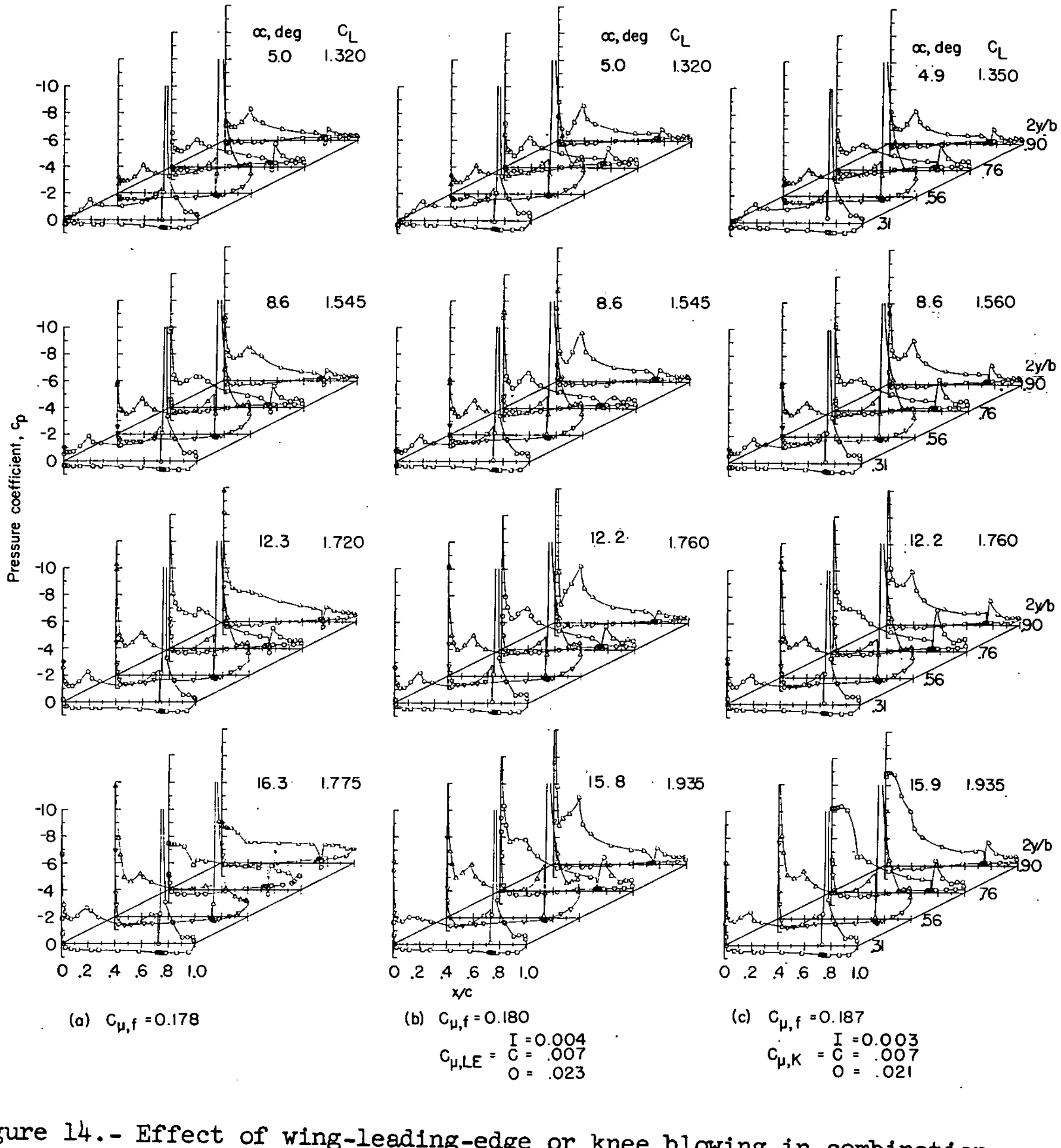


Figure 14.- Effect of wing-leading-edge or knee blowing in combination with flap blowing on chordwise pressure distributions. Wing leading edge drooped 40°; $\delta_f = 60°$; $\delta_a = 0°$.

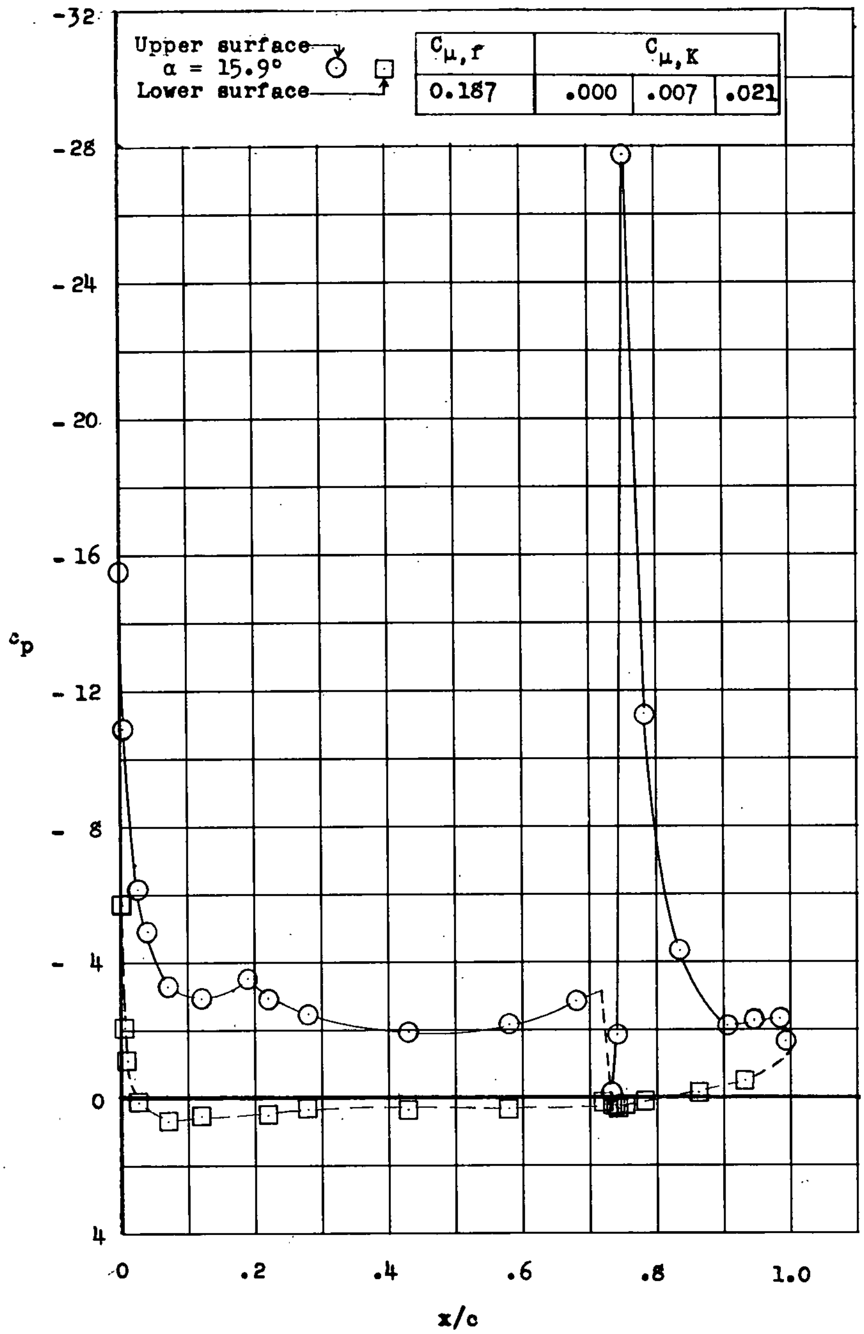


Figure 15.- Variation of chordwise pressure distribution at station 2 with knee blowing applied in combination with a high rate of flap blowing for the 40° drooped-leading-edge configuration. $\delta_f = 60^{\circ}$; $\delta_a = 0^{\circ}$.

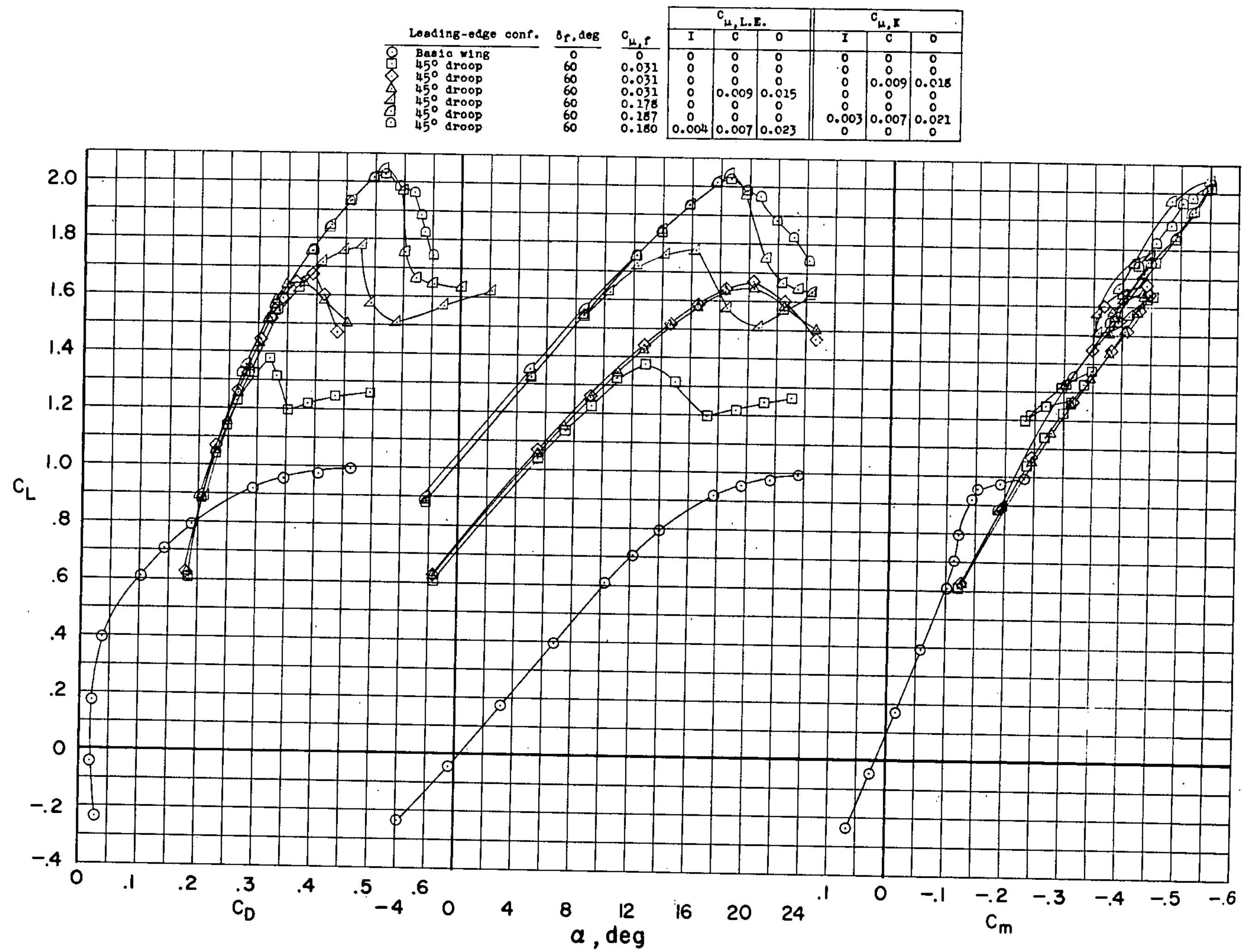
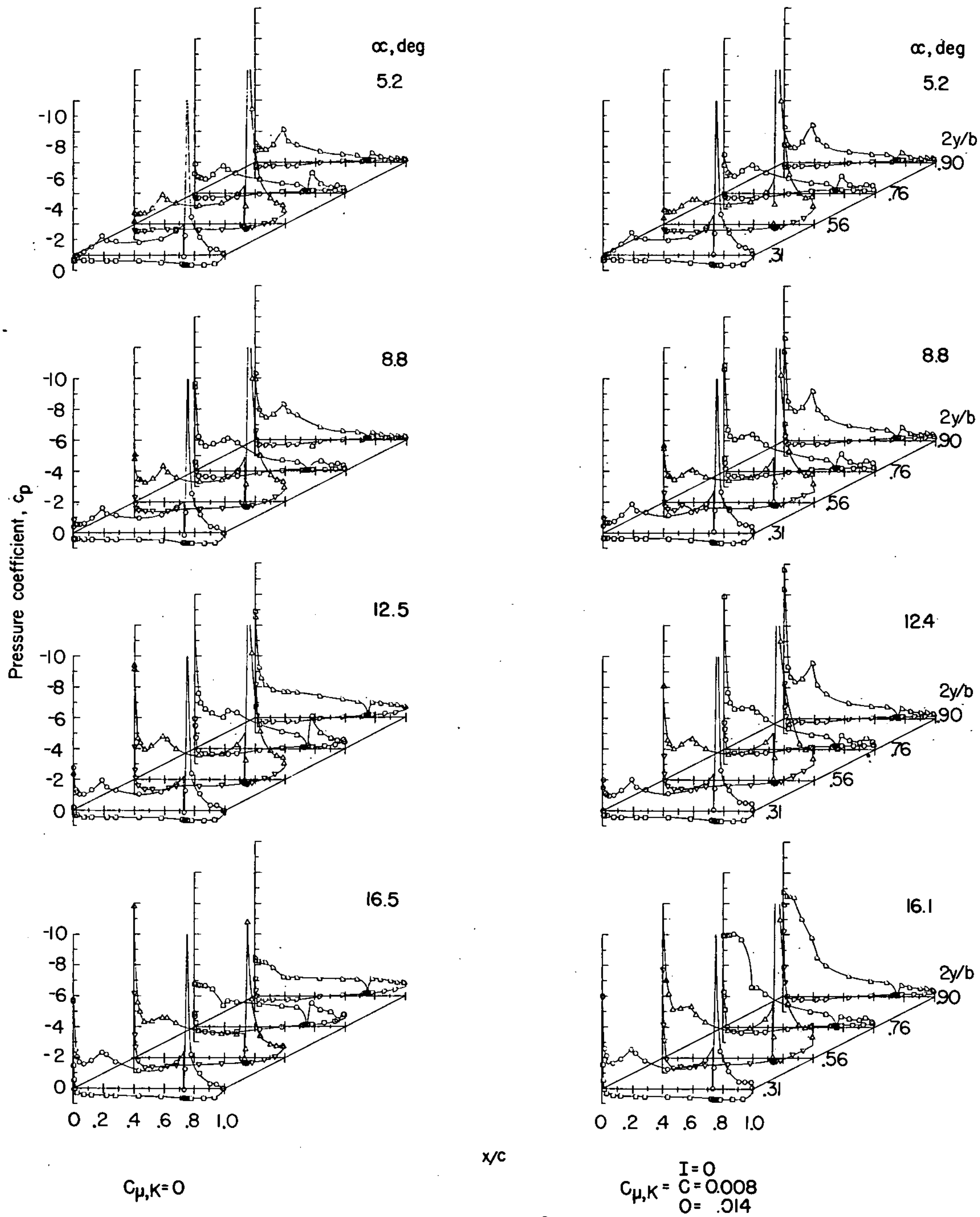
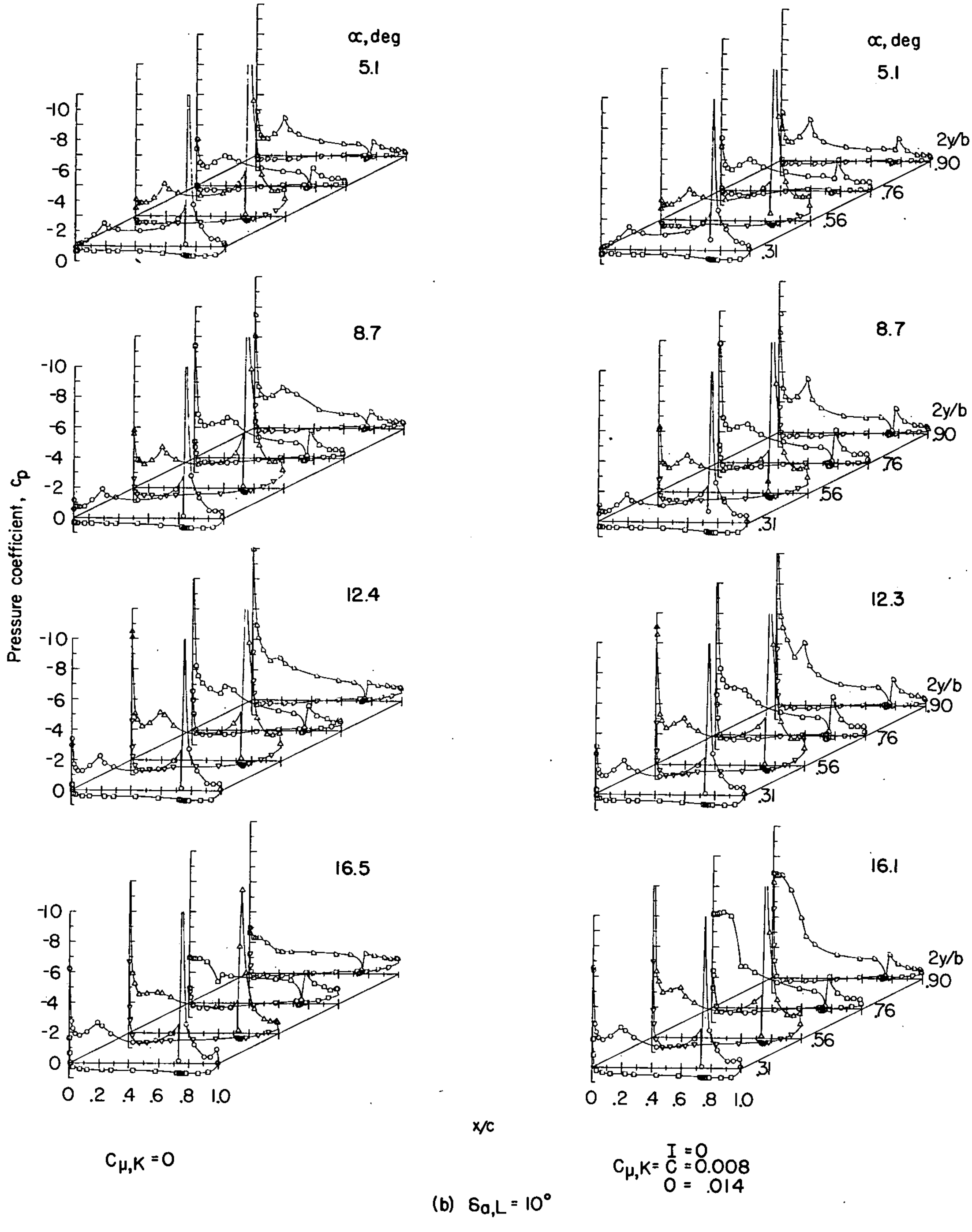


Figure 16.- Effect on aerodynamic characteristics of flap blowing alone and in combination with leading-edge or knee blowing. Leading edge drooped 40° and 45°. $\delta_f = 60°$; $\delta_a = 0°$.



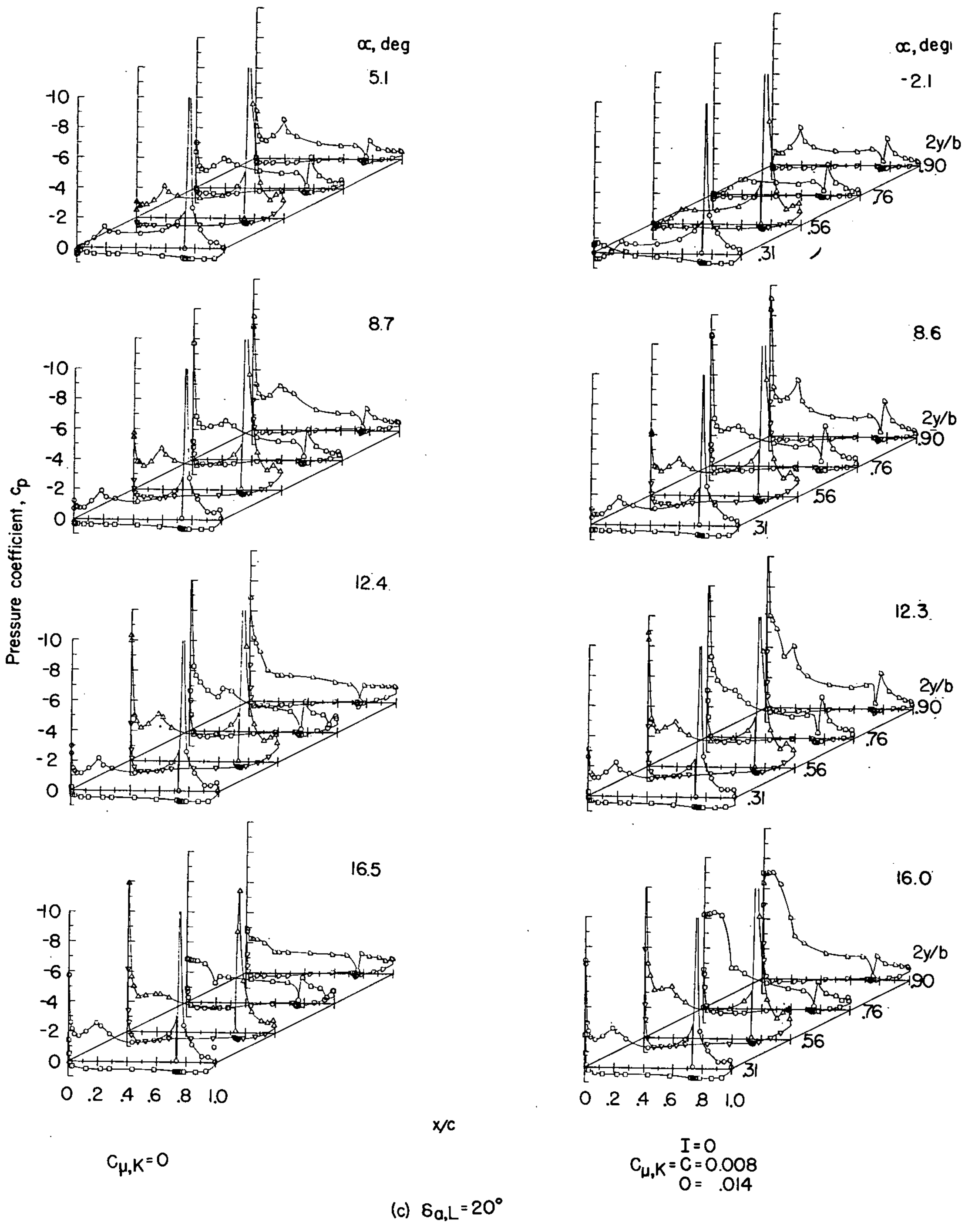
(a) $\delta_{a,L} = 0^{\circ}$

Figure 17.- Effect of knee blowing on chordwise pressure distributions with several aileron deflection angles for the 40° drooped-leading-edge configuration. $C_{\mu,f} \approx 0.100$; $\delta_f = 60^{\circ}$.



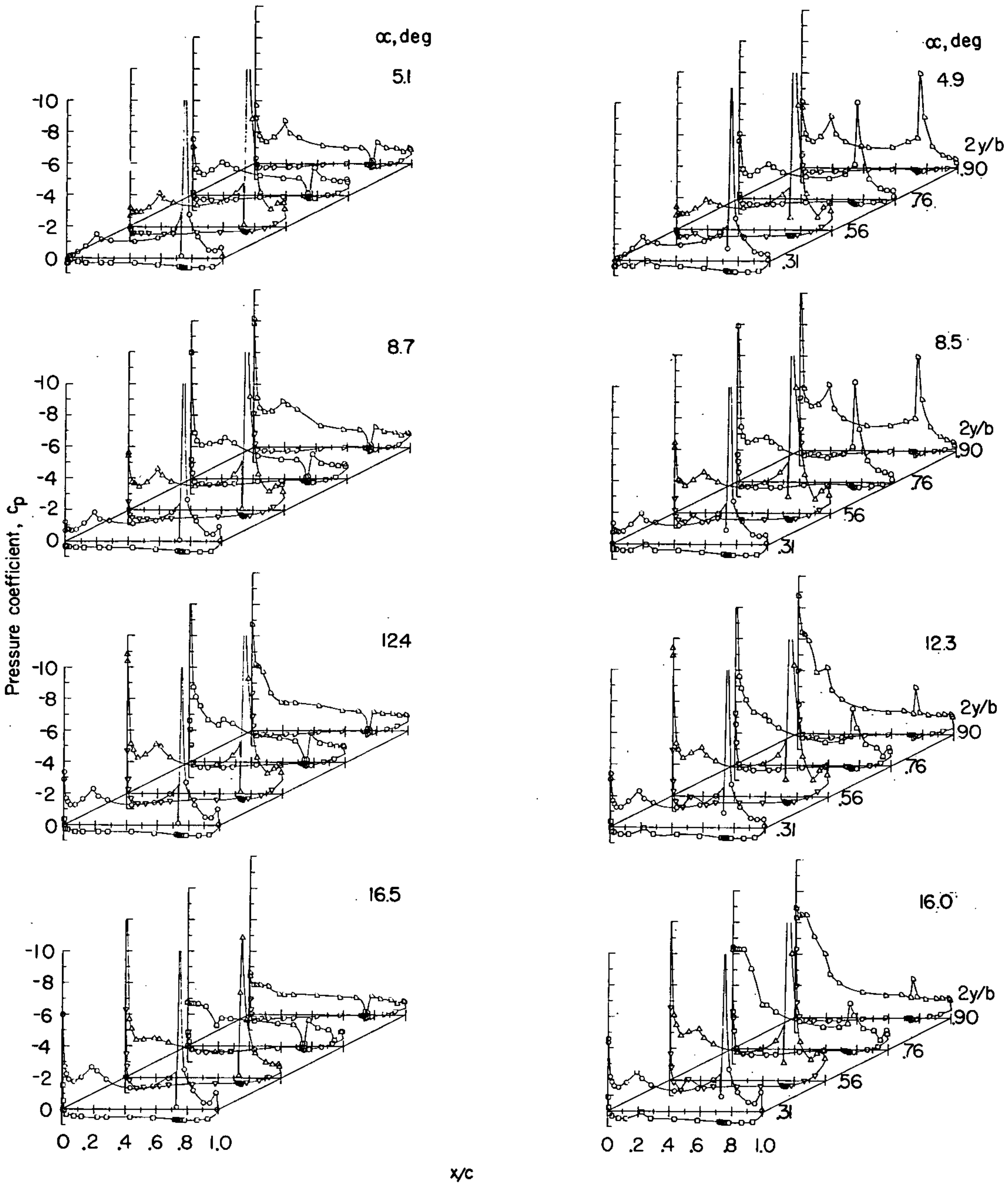
(b) $\delta_{a,L} = 10°$

Figure 17.- Continued.



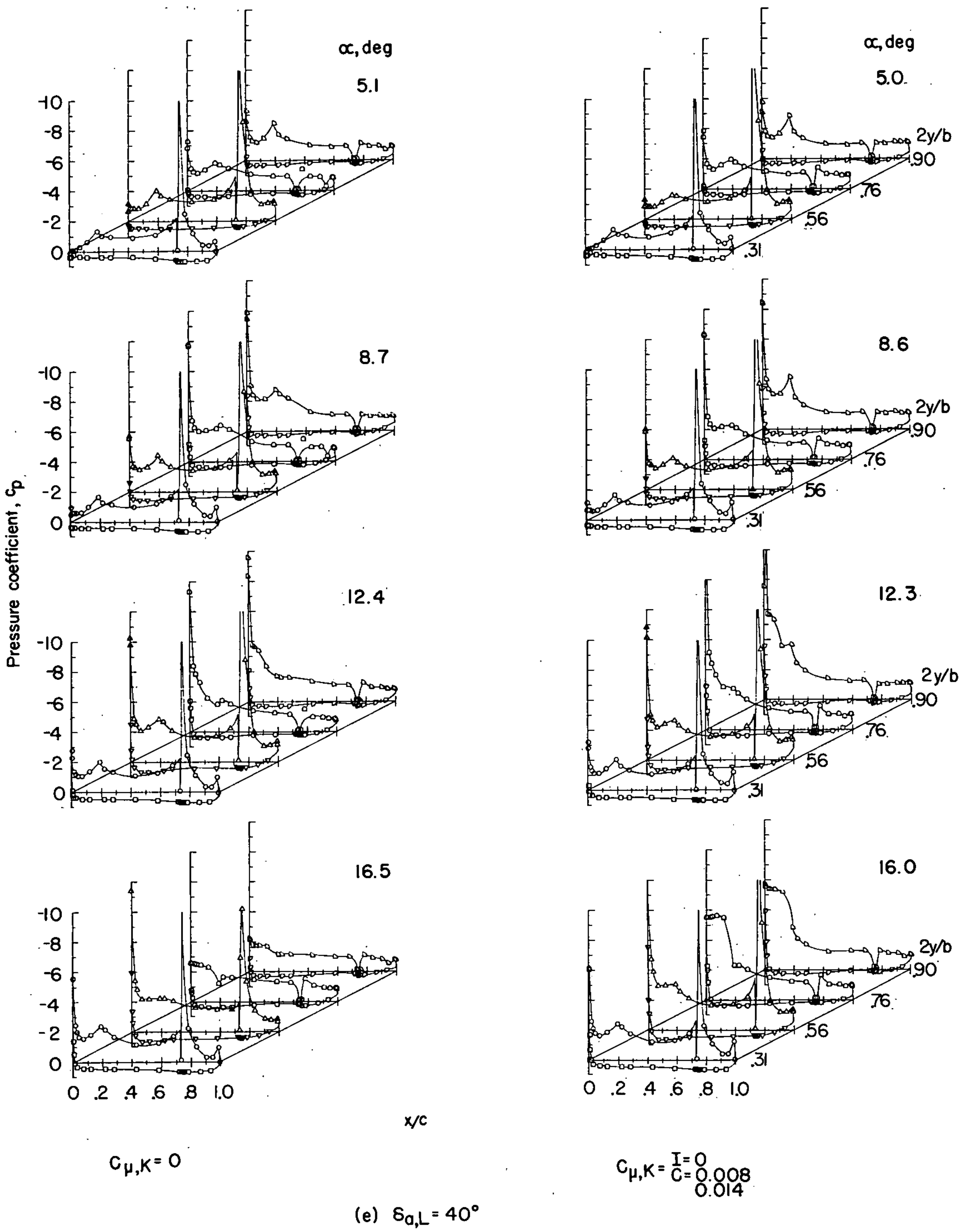
(c) $\delta_{a,L} = 20°$

Figure 17.- Continued.



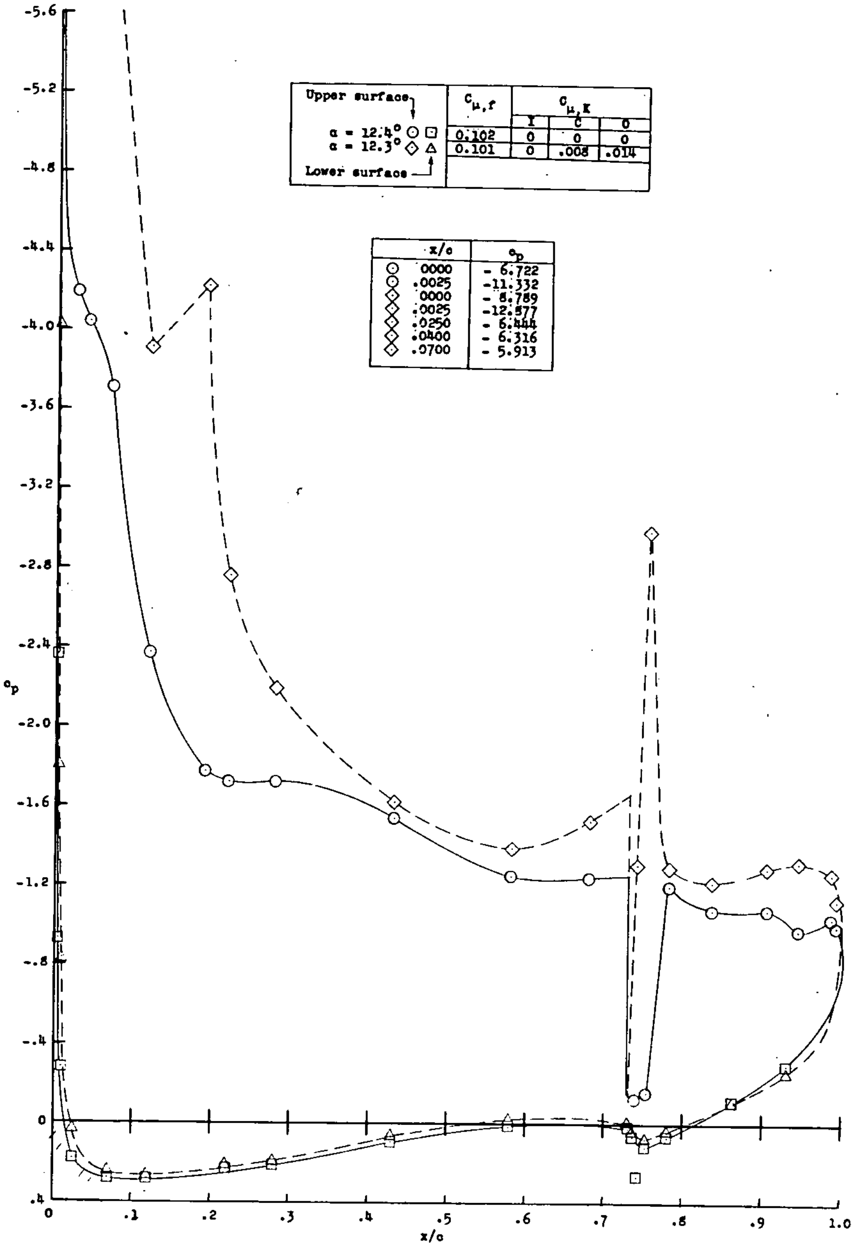
(d) $\delta_{a,L} = 30°$

Figure 17.- Continued.



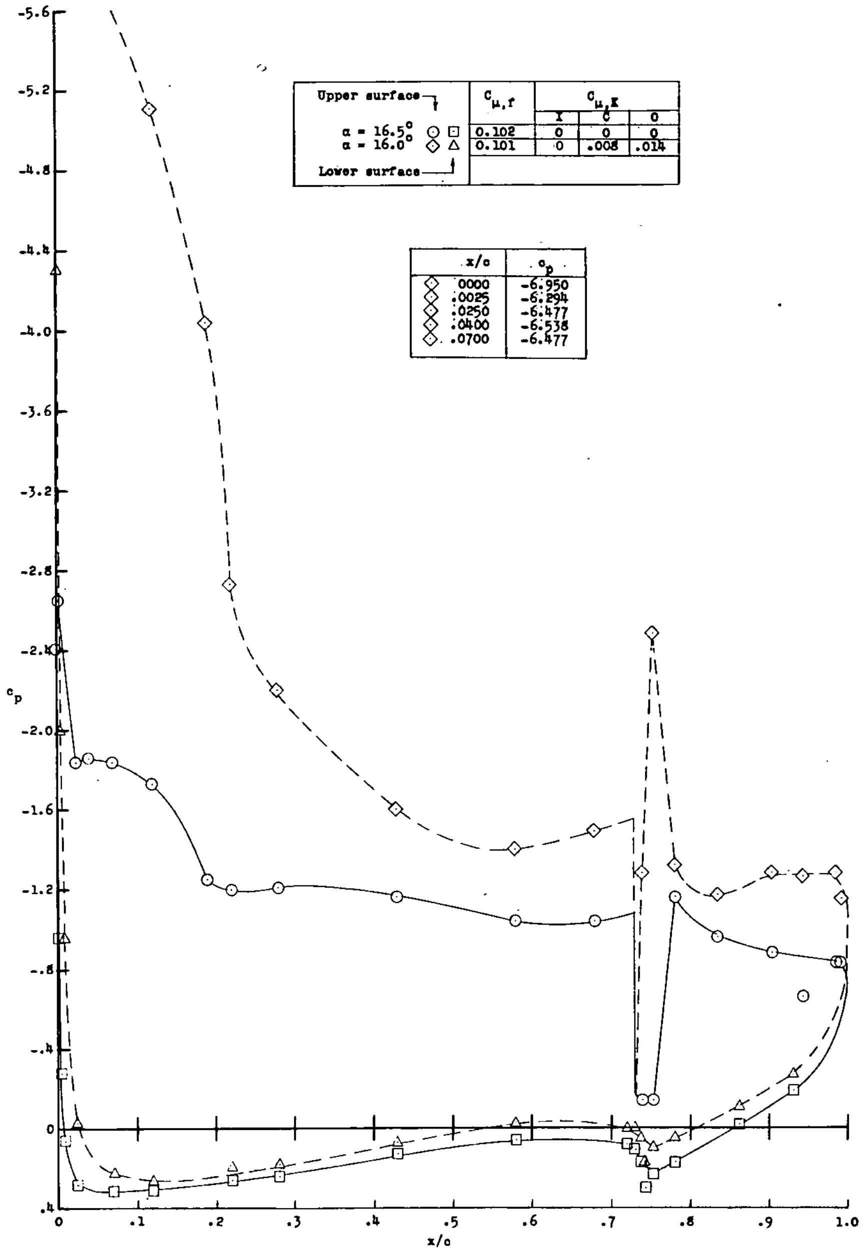
(e) $\delta_{a,L} = 40°$

Figure 17.- Concluded.



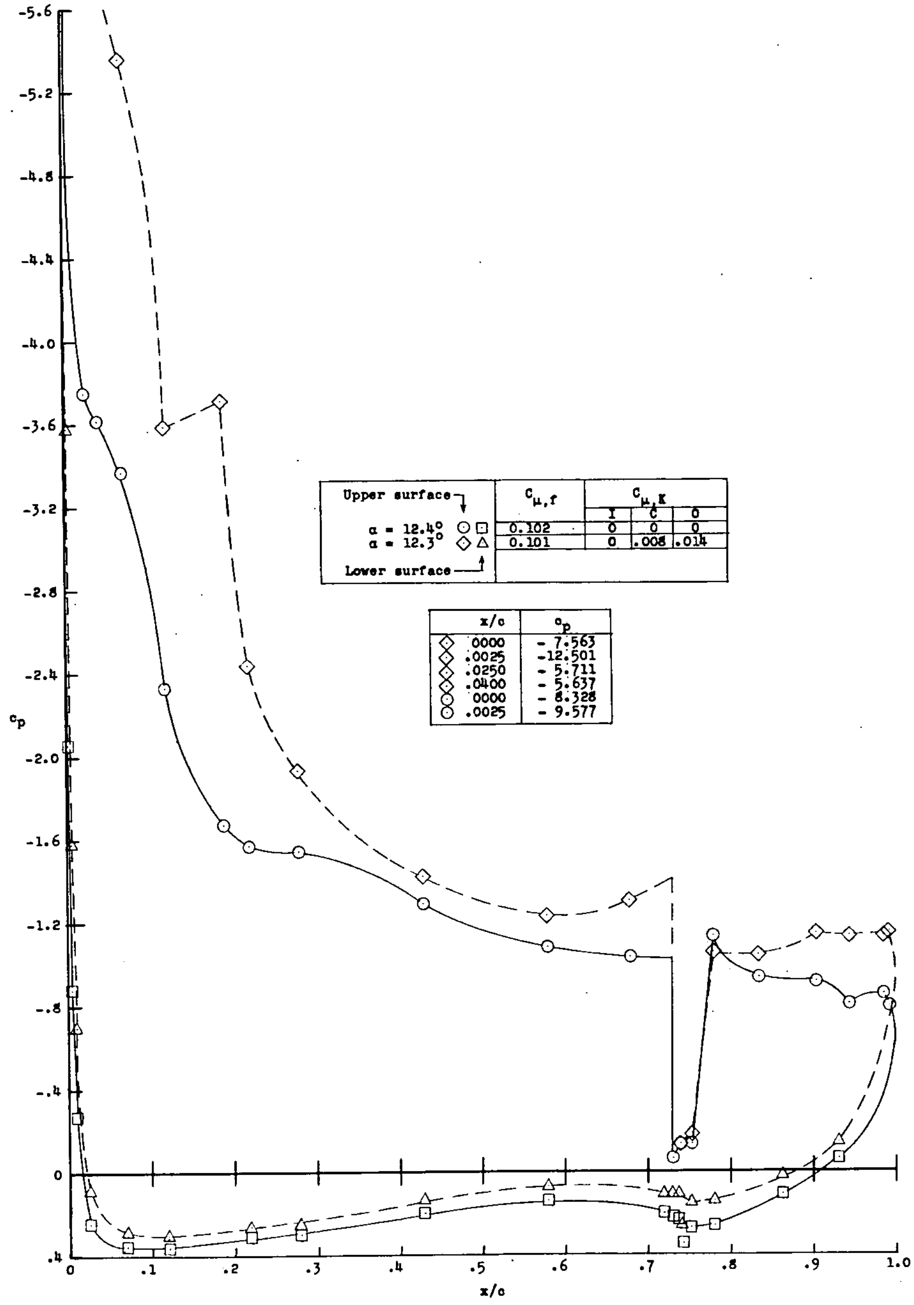
(a) $\alpha \approx 12^{\circ}$.

Figure 18.- Comparison of chordwise pressure distributions at station 4 without and with knee blowing applied in combination with flap blowing for the 40° drooped-leading-edge configuration. $\delta_f = 60^{\circ}$; $\delta_a = 30^{\circ}$.



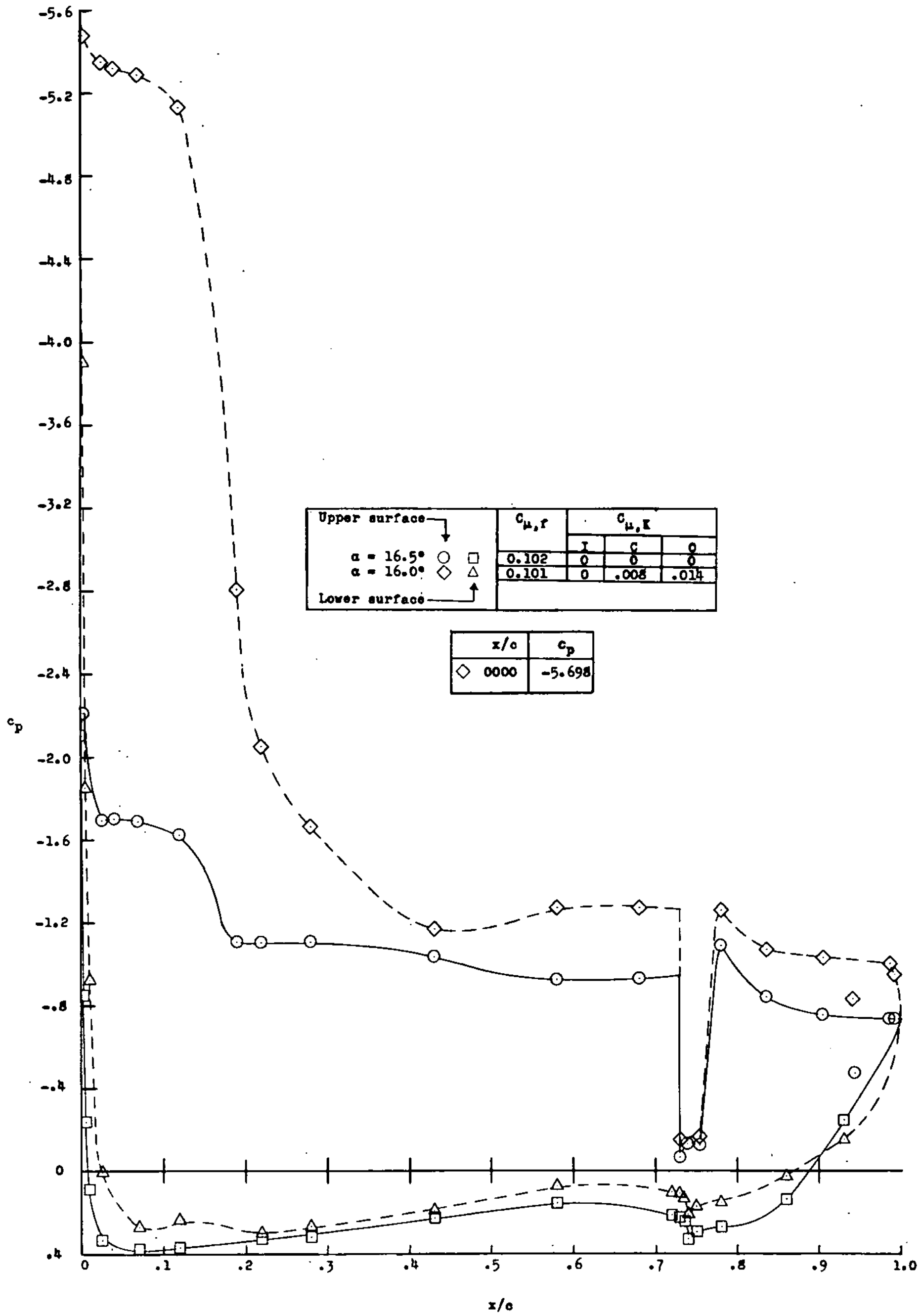
(b) $\alpha \approx 16^{\circ}$.

Figure 18.- Concluded.



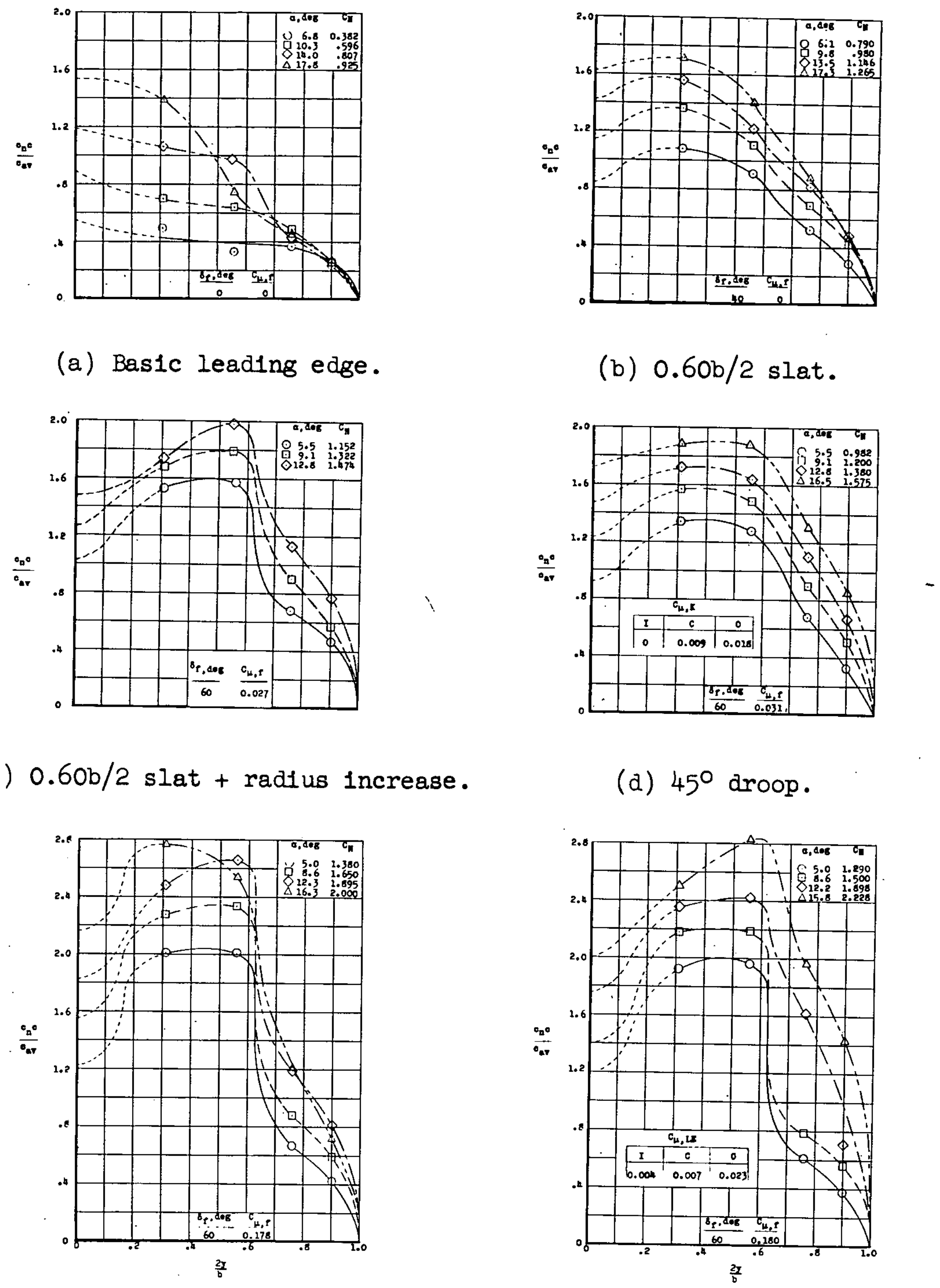
(a) $\alpha \approx 12^{\circ}$.

Figure 19.- Comparison of chordwise pressure distributions at station 4 without and with knee blowing applied in combination with flap blowing for the 40° drooped-leading-edge configuration. $\delta_f = 60^{\circ}$; $\delta_a = 40^{\circ}$.



(b) $\alpha \approx 16^{o}$.

Figure 19.- Concluded.



(a) Basic leading edge.

(b) 0.60b/2 slat.

(c) 0.60b/2 slat + radius increase.

(d) 45° droop.

(e) 40° droop.

(f) 40° droop.

Figure 20.- Span-loading characteristics of several wing configurations without and with leading-edge, knee, and flap blowing applied.

Leading-edge conf.	δ_f, deg	$C_{\mu,f}$	$C_{\mu,LE}$ I	$C_{\mu,LE}$ C	$C_{\mu,LE}$ O	$C_{\mu,K}$ I	$C_{\mu,K}$ C	$C_{\mu,K}$ O
Basic wing	0	0	0	0	0	0	0	0
0.60b/2 Slat	40	0	0	0	0	0	0	0
0.60b/2 Slat + radius increase	60	0.027	0	0	0	0	0	0
45° Droop	60	0.031	0	0	0	0	0.009	0.018
40° Droop	60	0.178	0	0	0	0	0	0
40° Droop	60	0.180	0.0004	0.007	0.023	0	0	0

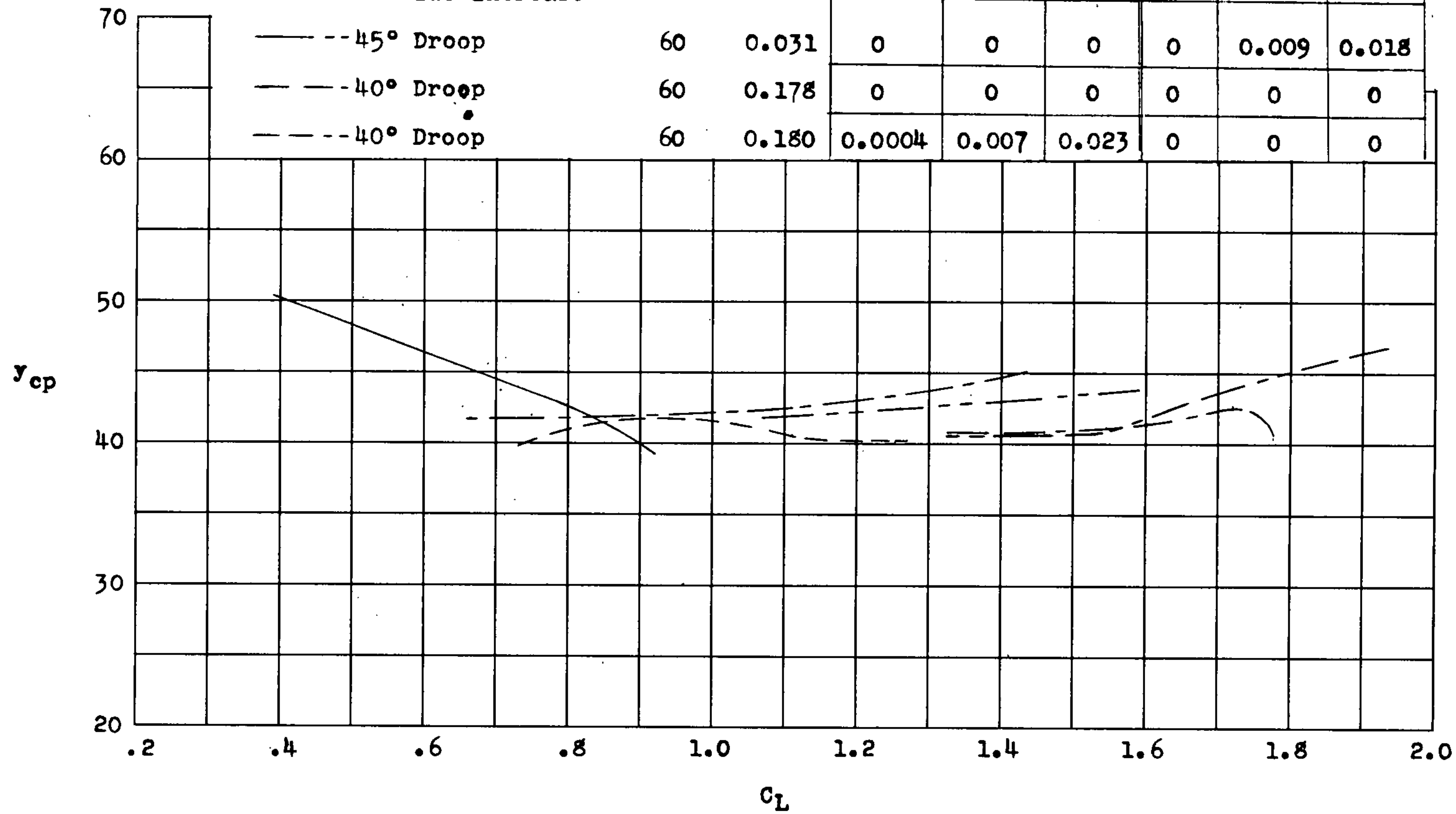
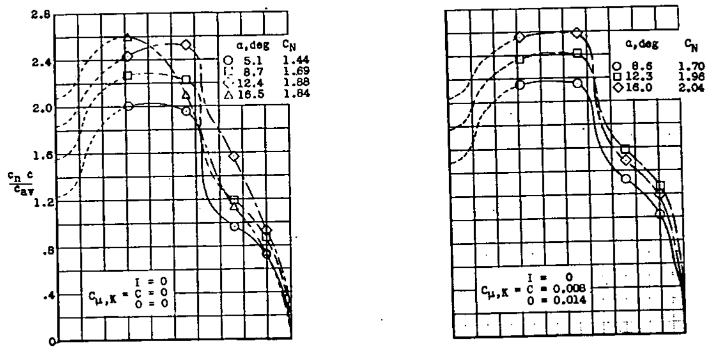
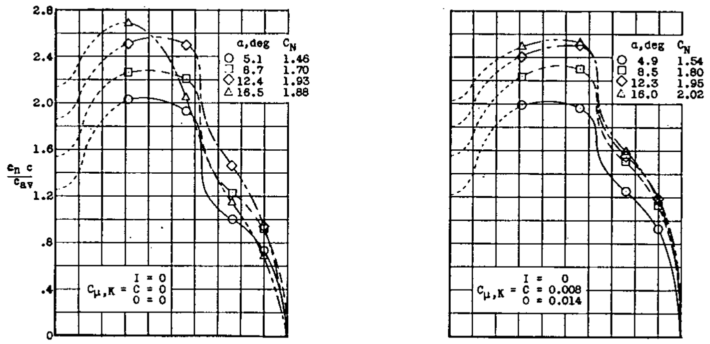


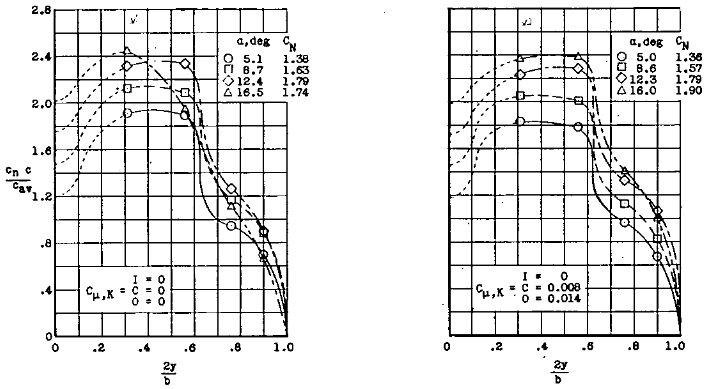
Figure 21.- Variation of the lateral center of pressure for several wing configurations.



(a) $\delta_a = 20^{\circ}$.



(b) $\delta_a = 30^{\circ}$.



(c) $\delta_a = 40^{\circ}$.

Figure 22.- Span-loading characteristics of the 40° drooped-leading-edge configuration without and with knee blowing applied, for several aileron deflection angles. $C_{\mu,f} \approx 0.10$; $\delta_f = 60^{\circ}$.

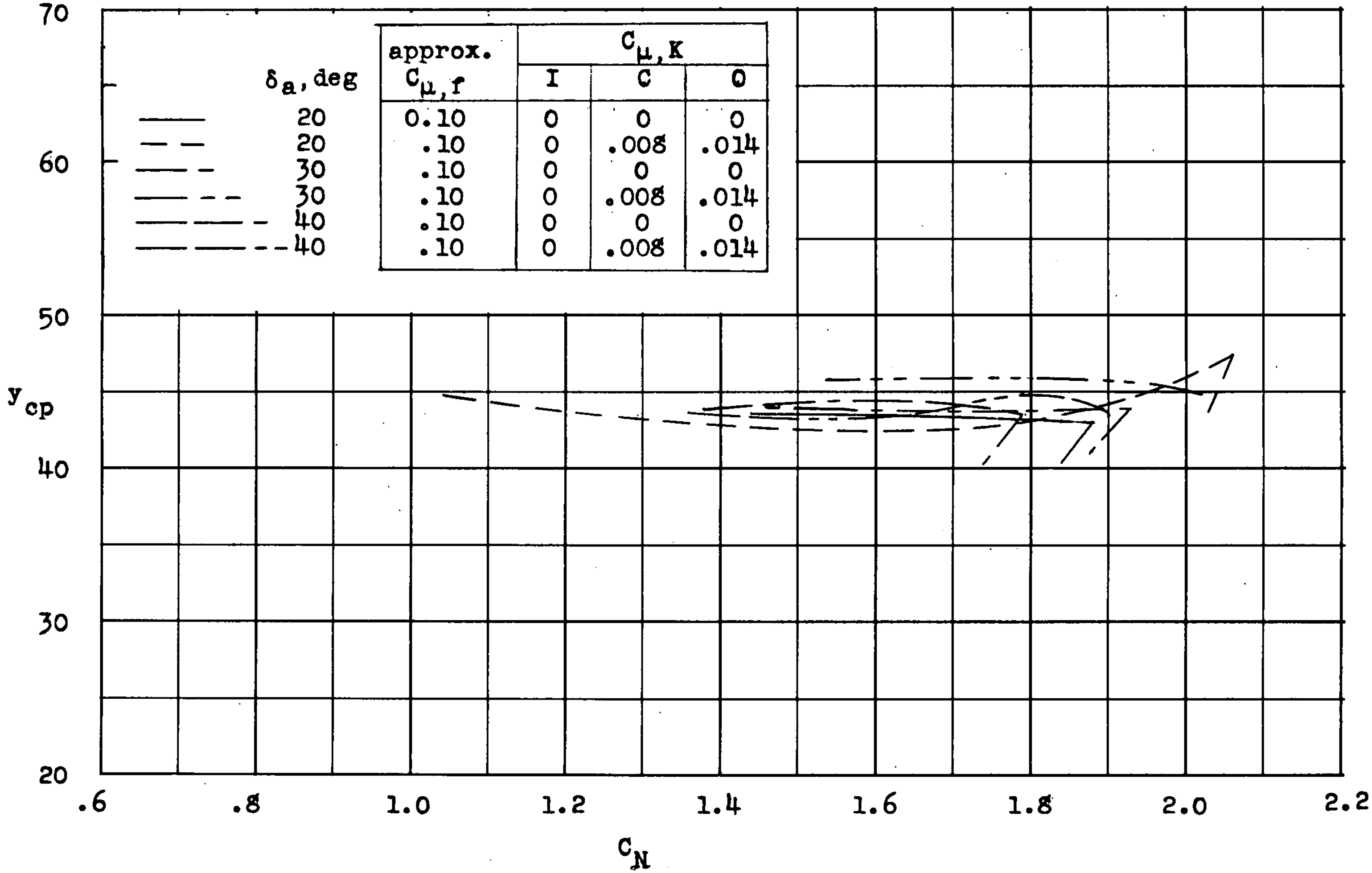


Figure 23.- Variation of the lateral center of pressure of the 40° drooped-leading-edge configuration without and with knee blowing applied in combination with flap blowing, for several aileron deflection angles. $\delta_f = 60^\circ$.

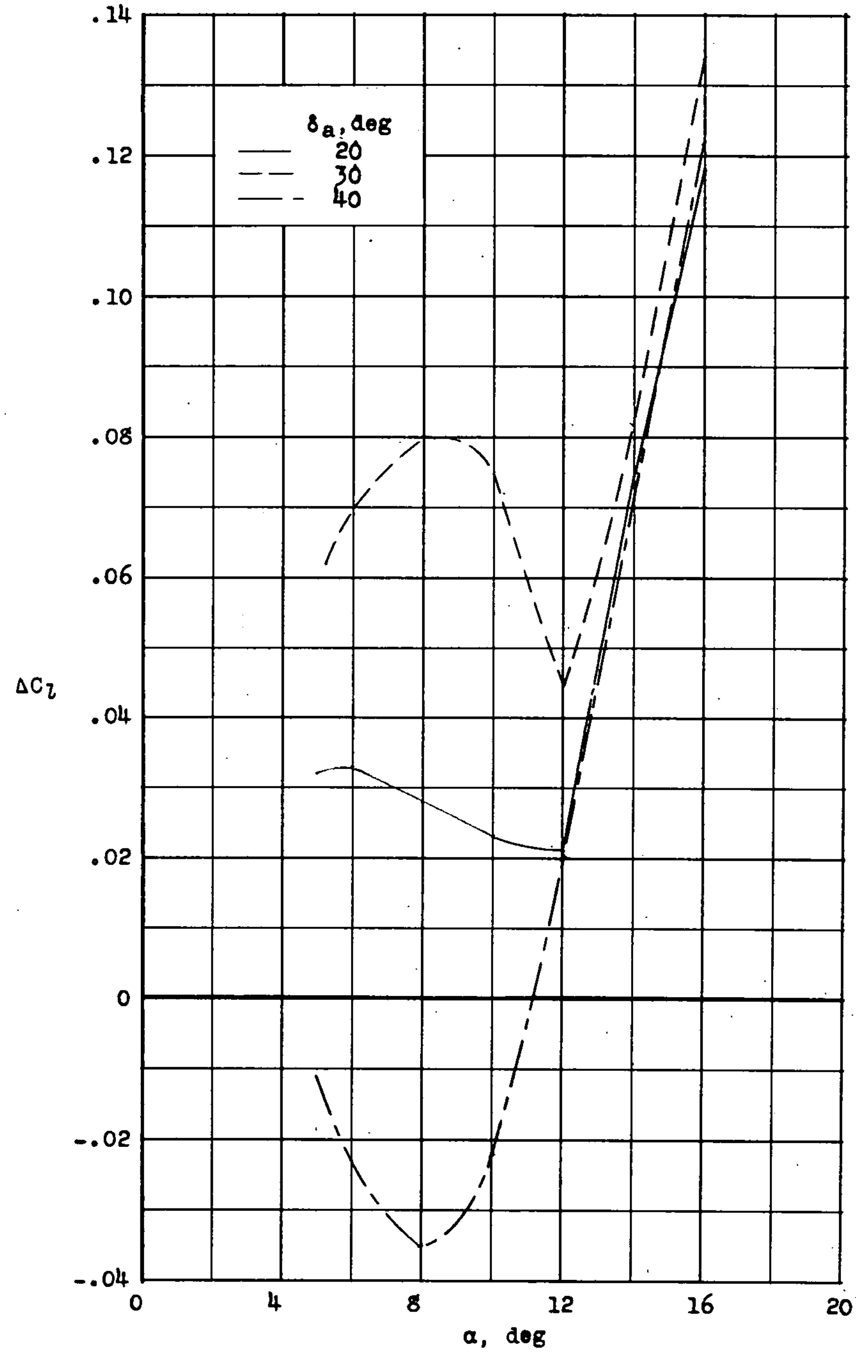
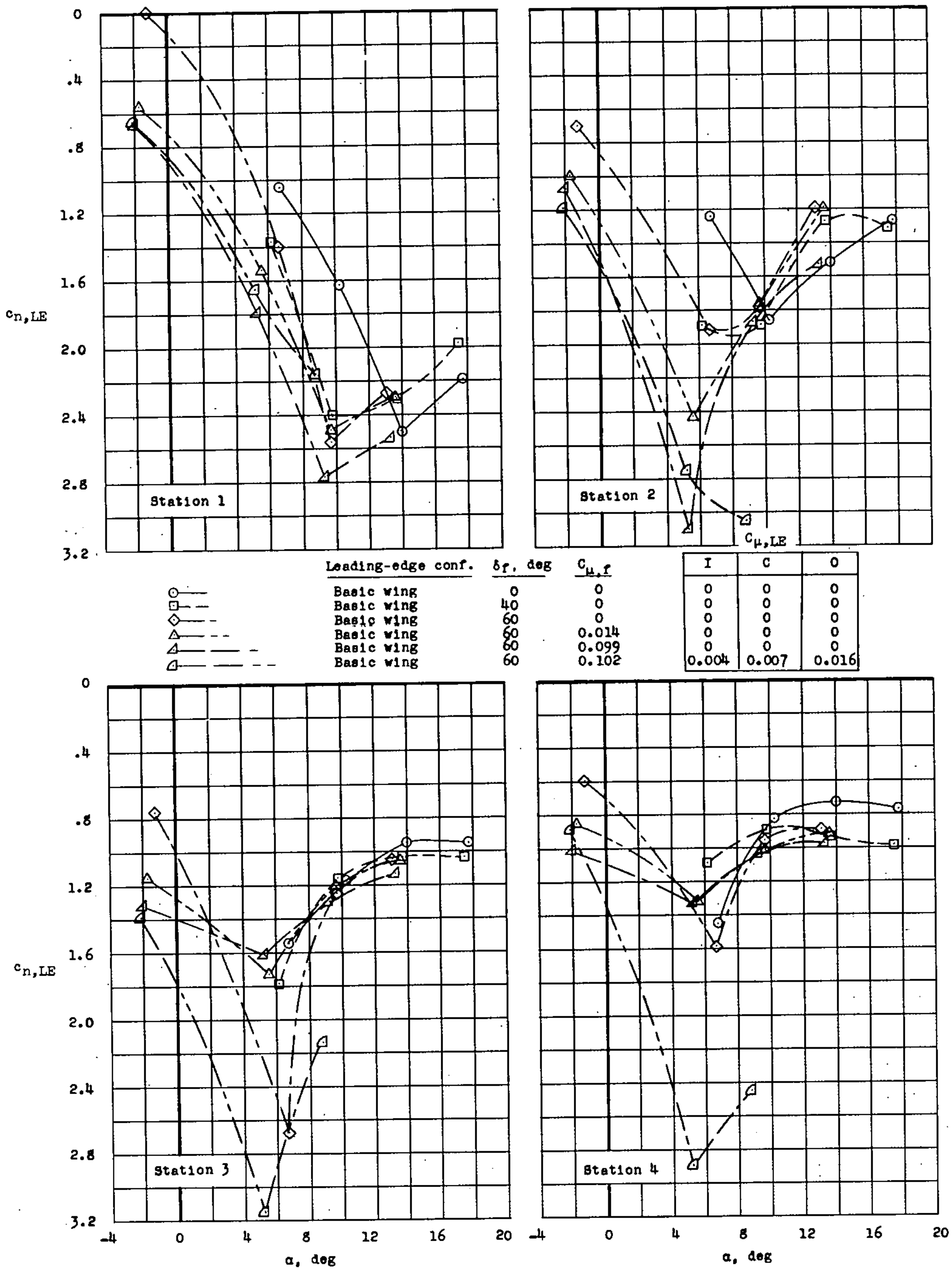
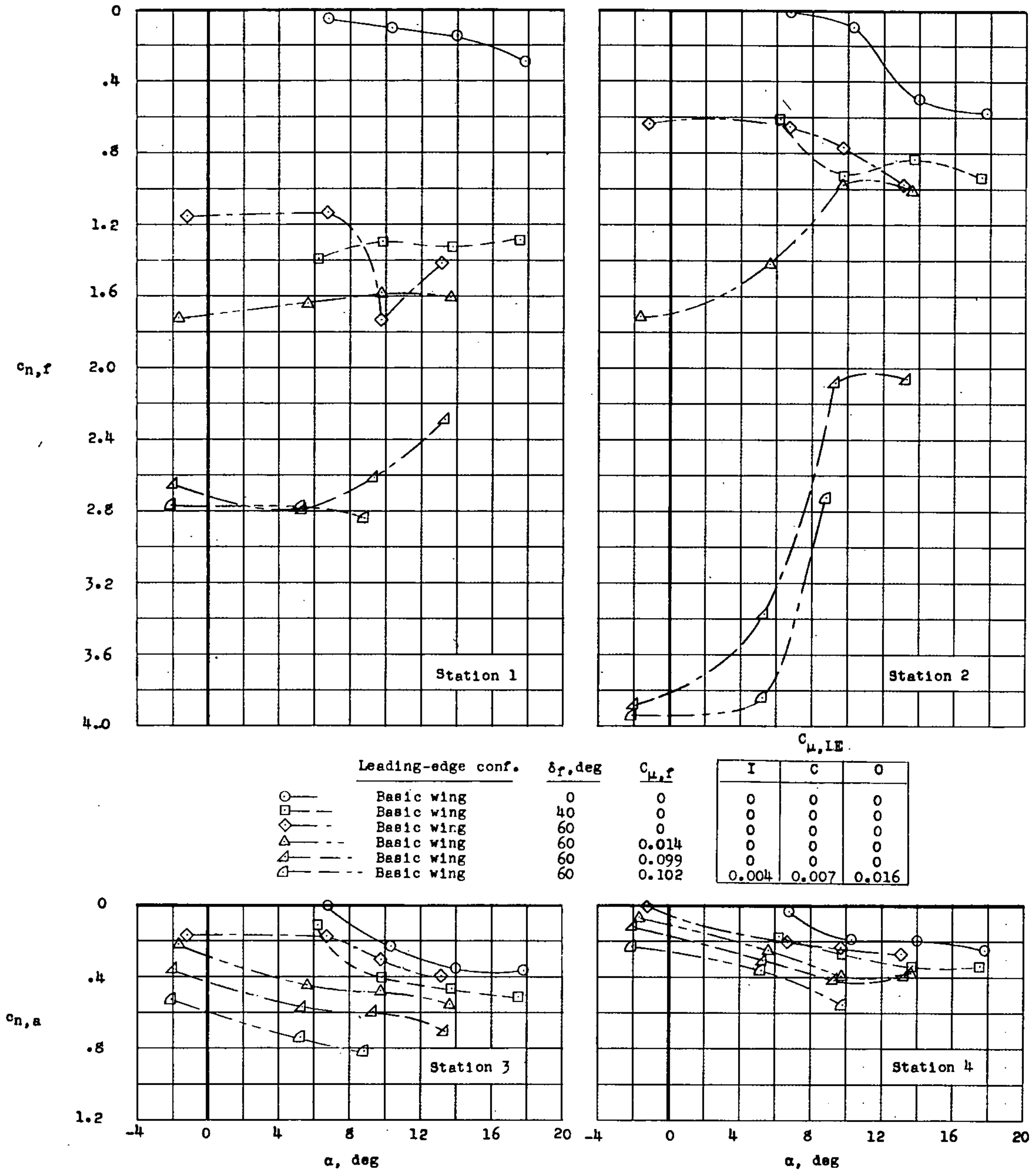


Figure 24.- Variation of the increment of rolling-moment coefficient produced by shutting off the knee blowing on one wing. 40° drooped-leading-edge configuration with flap blowing applied. $\delta_f = 60°$; $C_{\mu,f} \approx 0.10$; $C_{\mu,K}$: I = 0, C = 0.008, O = 0.014.



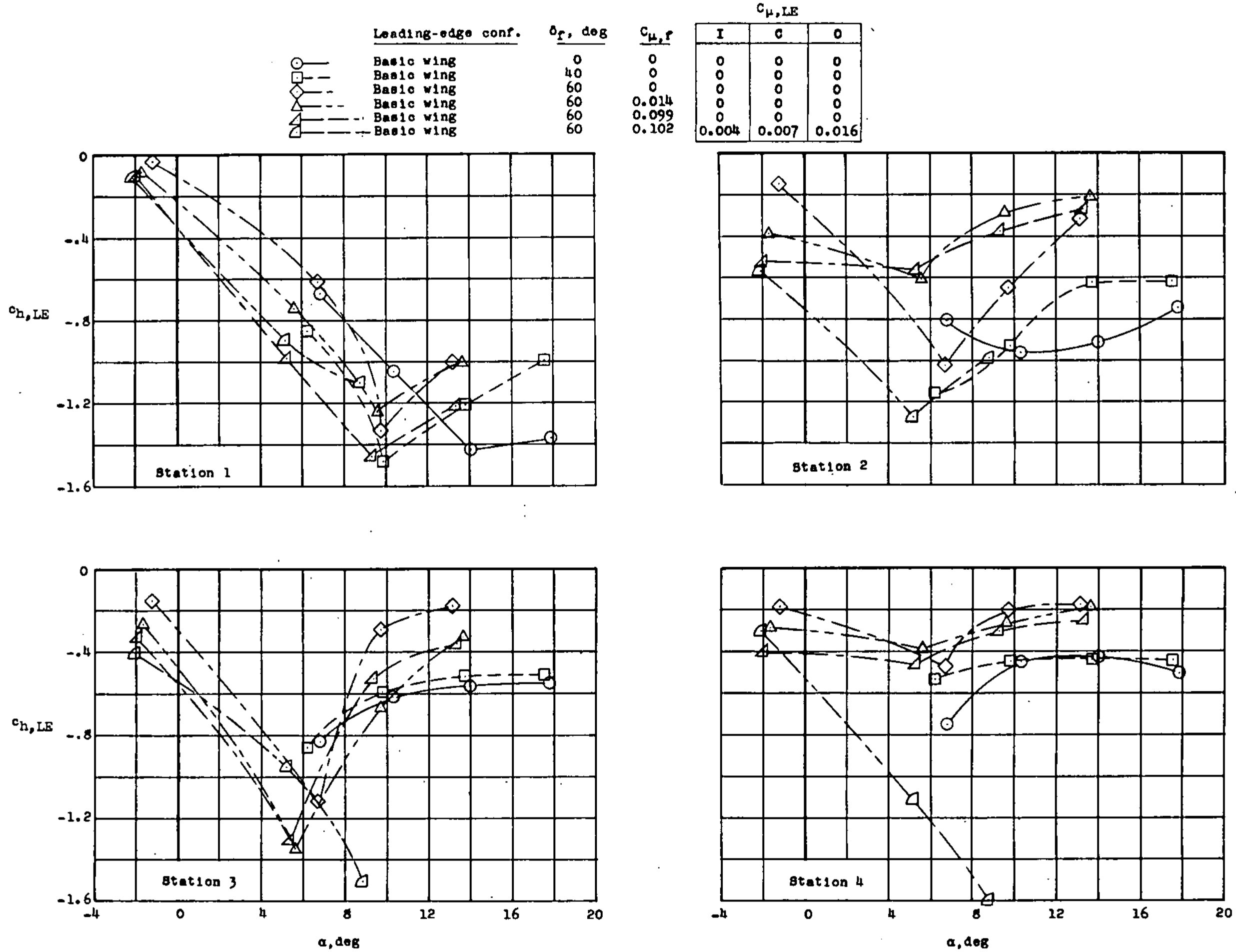
(a) Leading edge.

Figure 25.- Variation of leading-edge, flap, and aileron section normal-force coefficients with angle of attack, without and with leading-edge and flap blowing applied. $\delta_a = 0^{\circ}$.



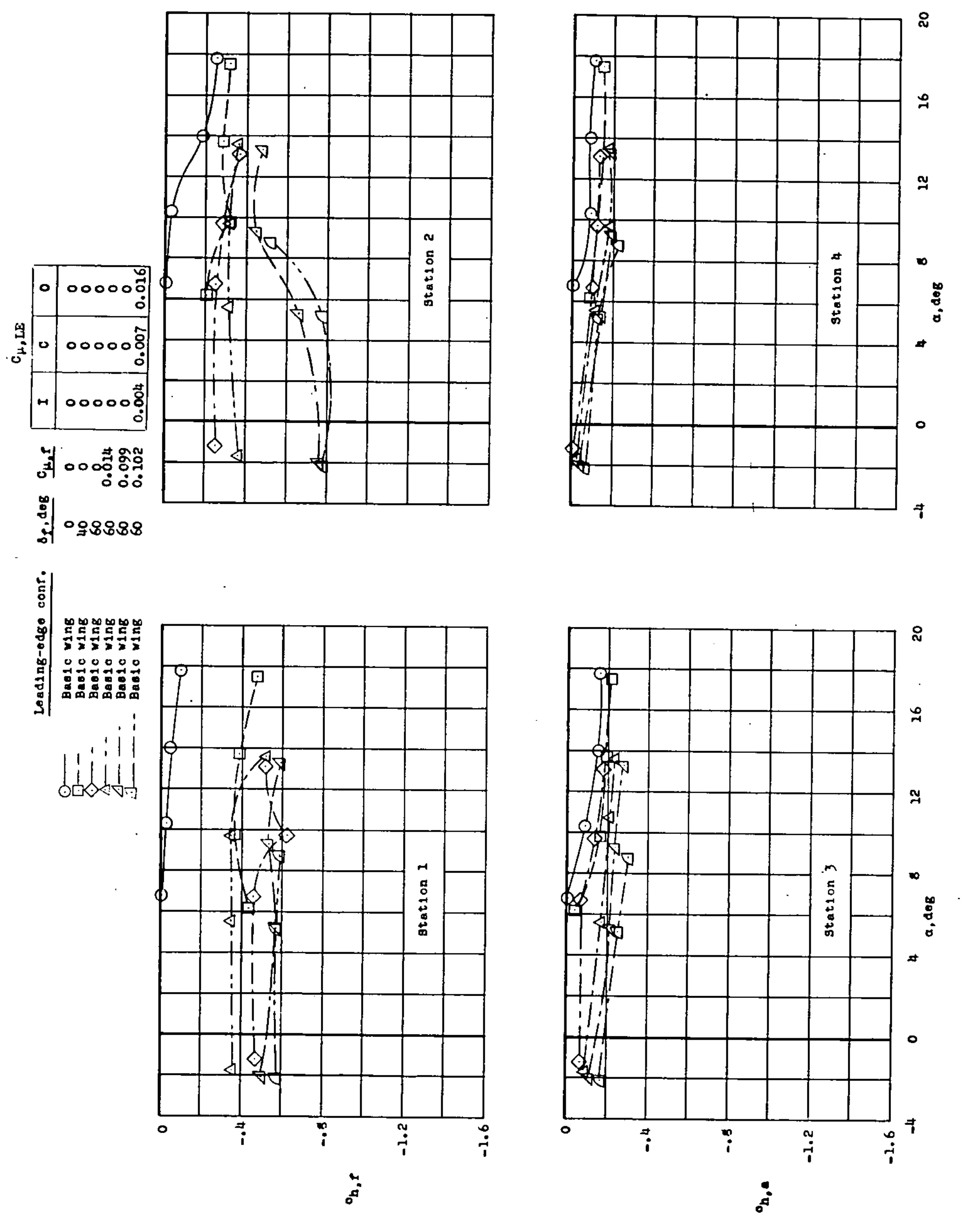
(b) Flap and aileron.

Figure 25.- Concluded.



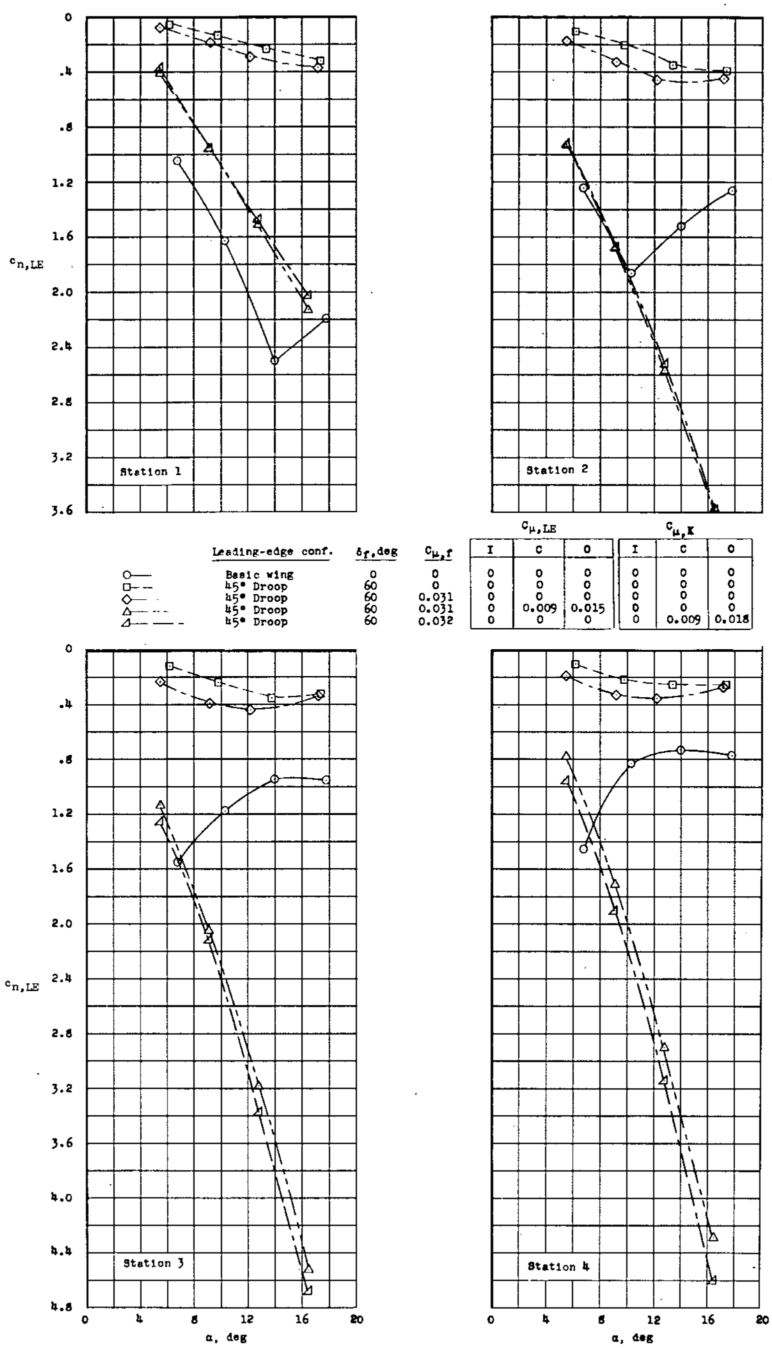
(a) Leading edge.

Figure 26.- Variation of leading-edge, flap, and aileron section hinge-moment coefficients with angle of attack, without and with leading-edge and flap blowing applied. $\delta_a = 0^{\circ}$.



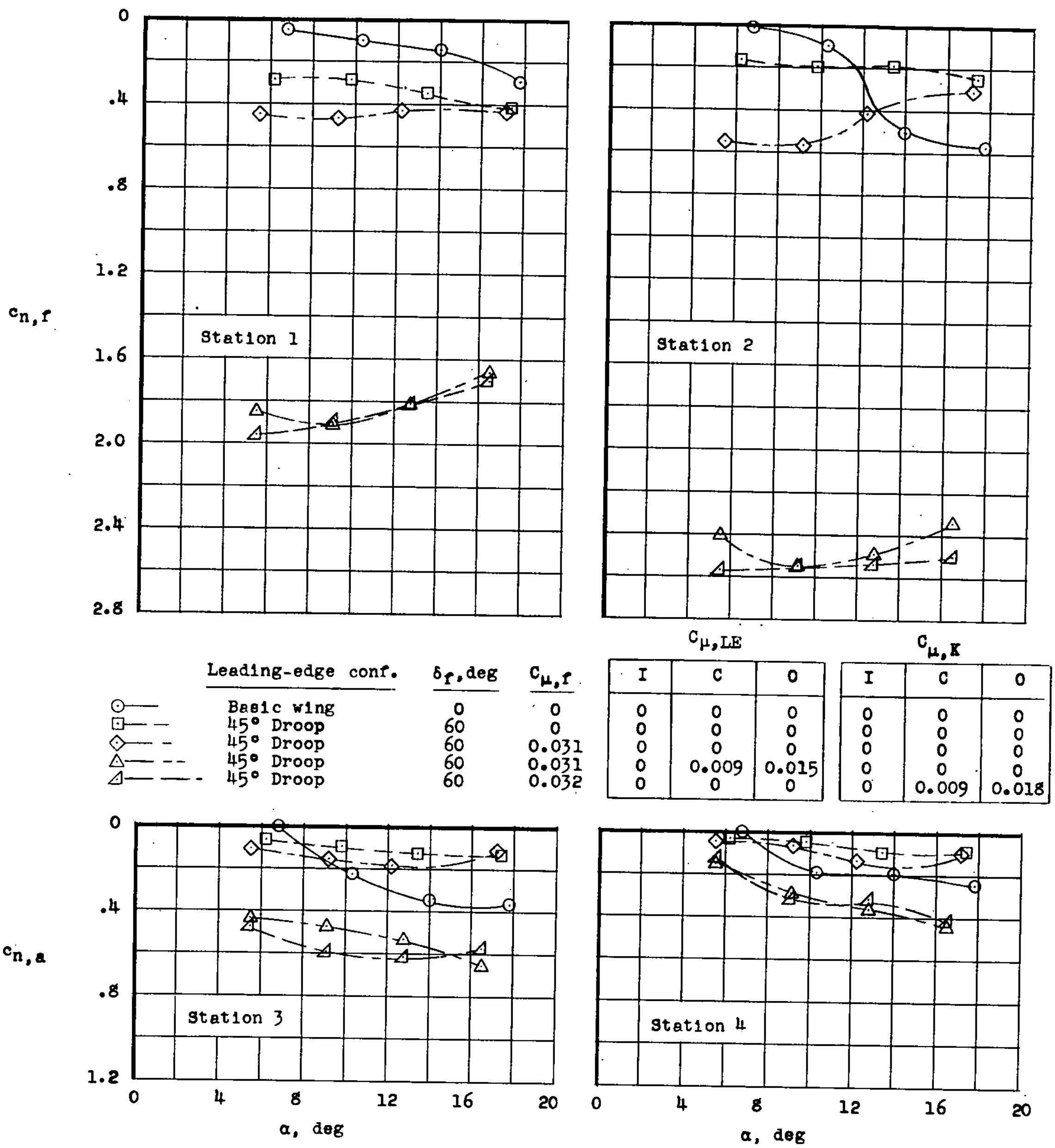
(b) Flap and aileron.

Figure 26.- Concluded.



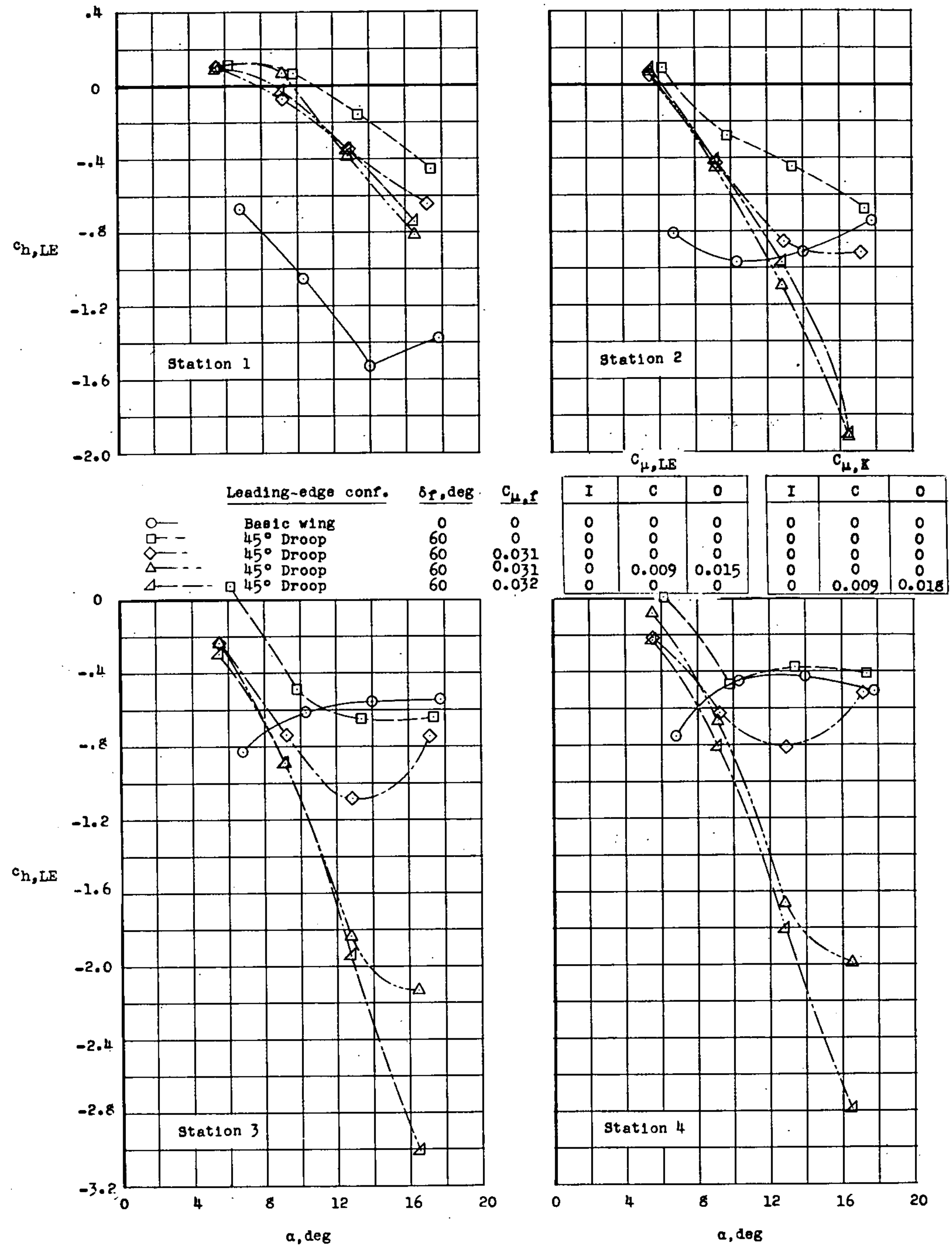
(a) Leading edge.

Figure 27.- Variation of leading-edge, flap, and aileron section normal-force coefficients with angle of attack, with leading-edge and flap blowing applied. $\delta_a = 0^{\circ}$.



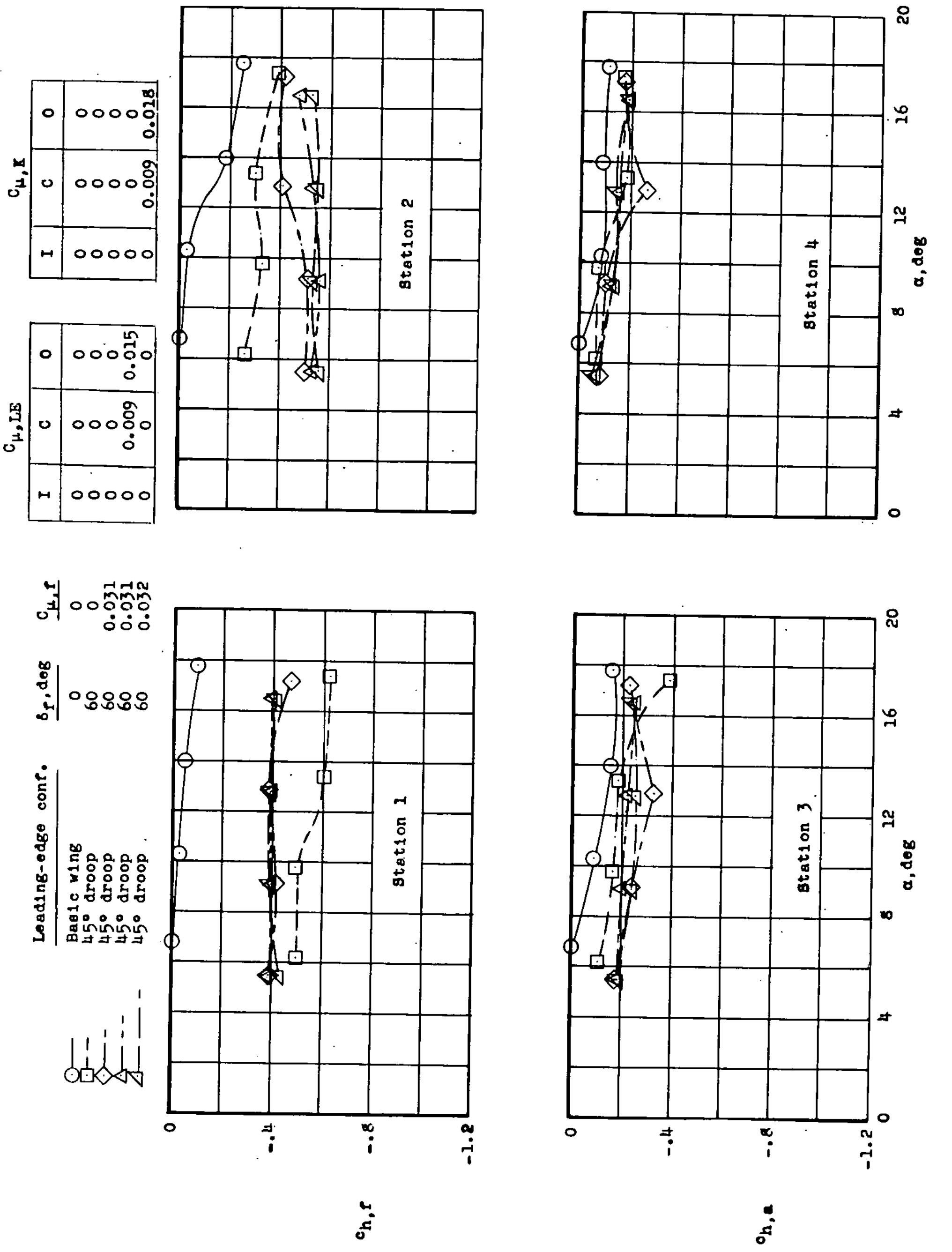
(b) Flap and aileron.

Figure 27.- Concluded.



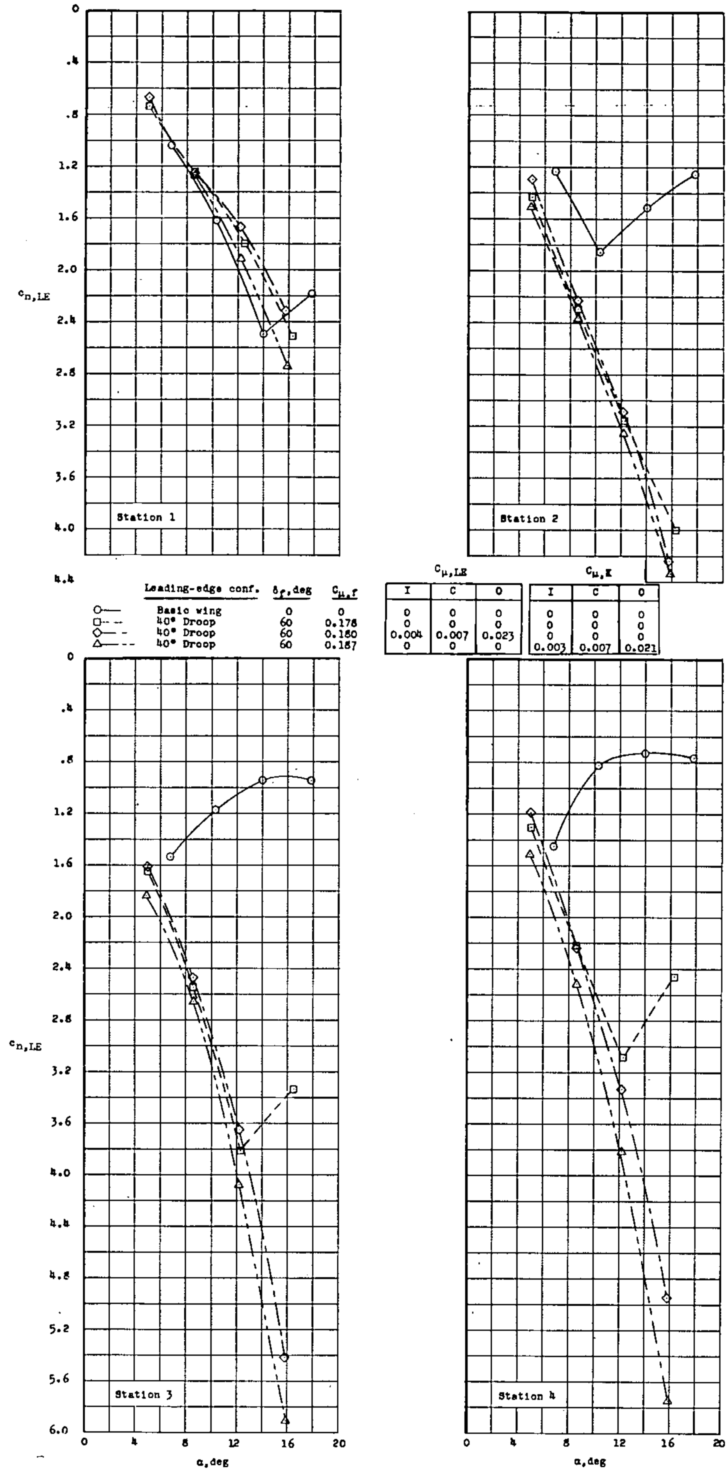
(a) Leading edge.

Figure 28.- Variation of leading-edge, flap, and aileron section hinge-moment coefficients with angle of attack, with leading-edge and flap blowing applied. $\delta_a = 0^{\circ}$.



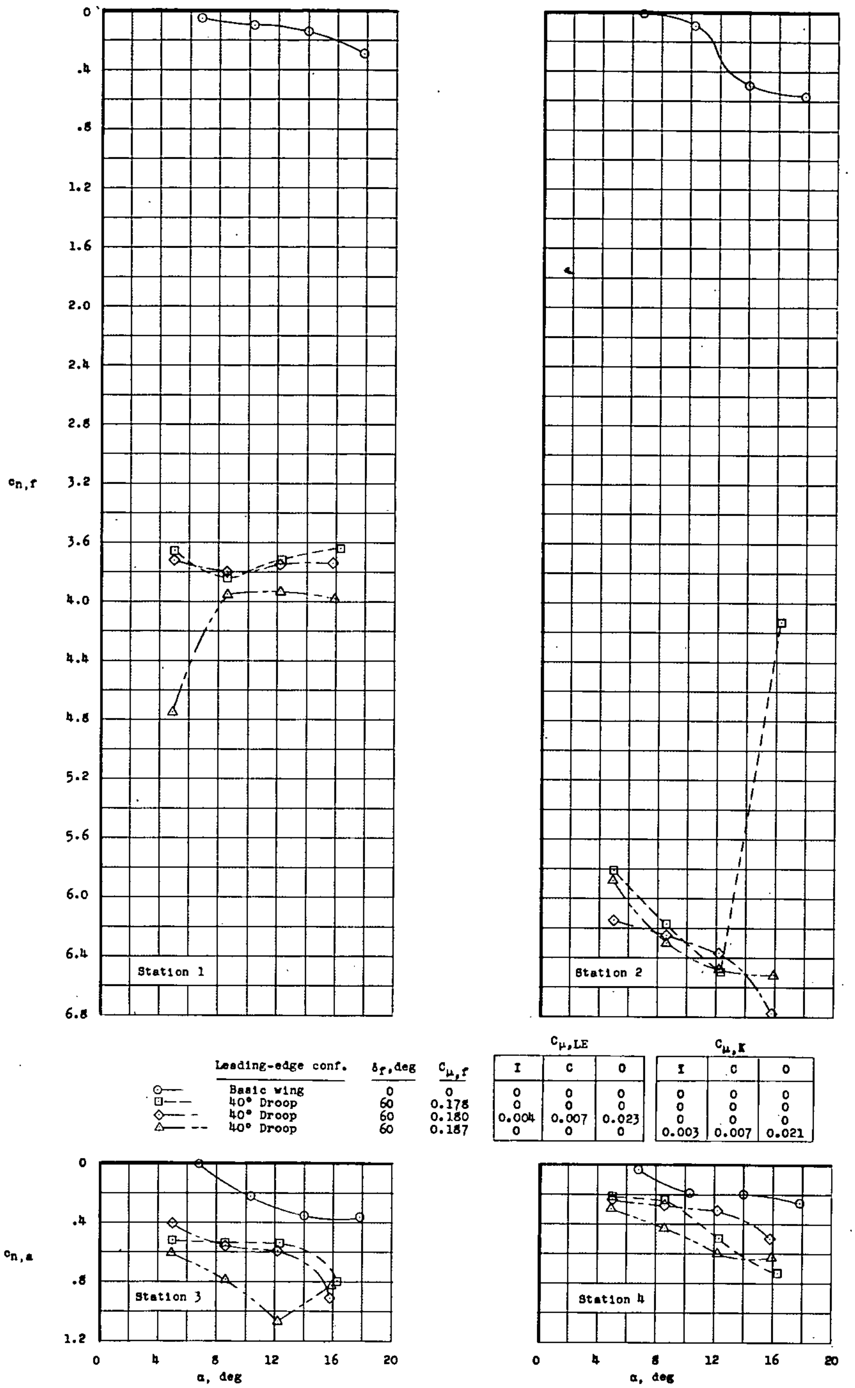
(b) Flap and aileron.

Figure 28.- Concluded.



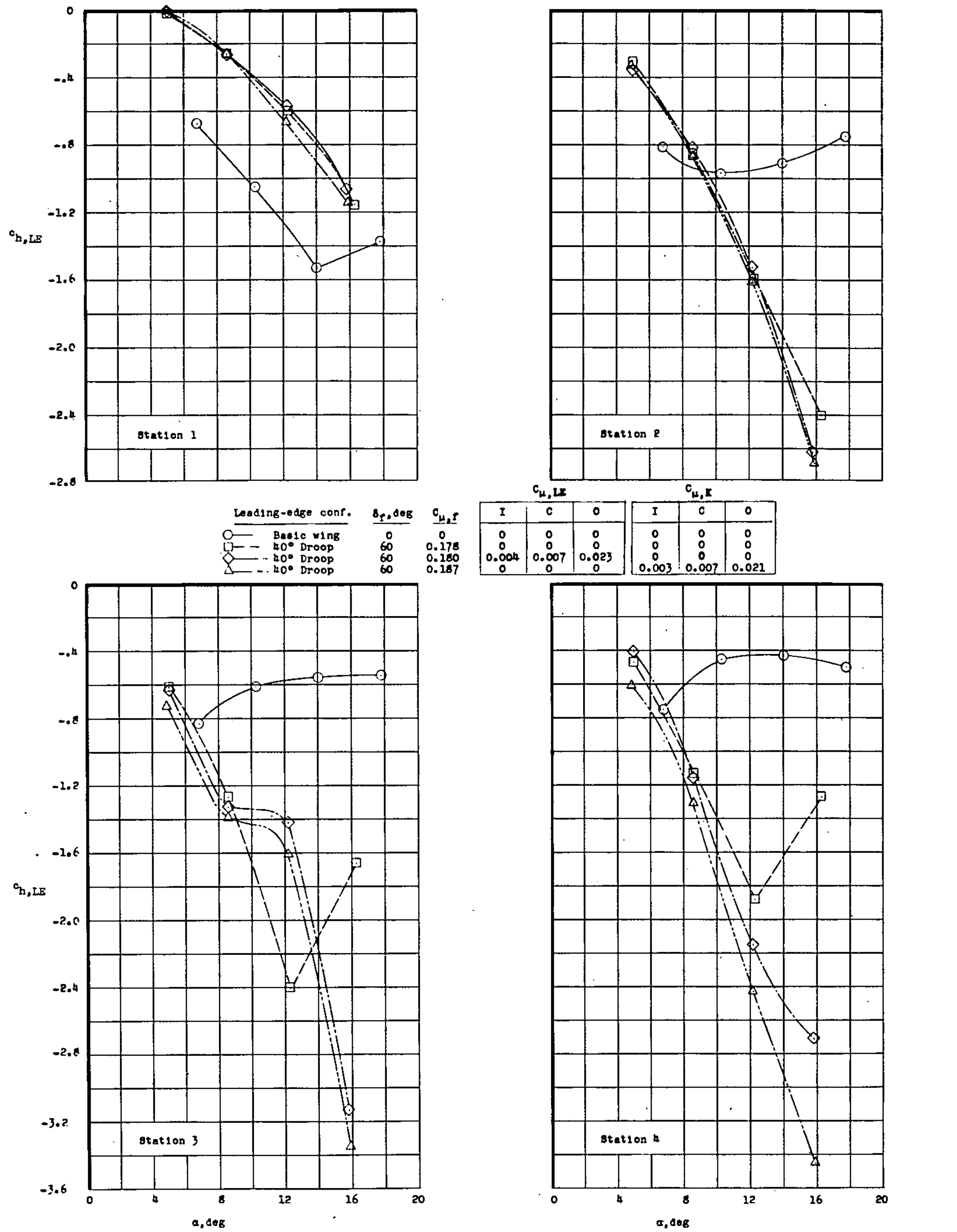
(a) Leading edge.

Figure 29.- Variation of leading-edge, flap, and aileron normal-force coefficients with angle of attack, without and with leading-edge and flap blowing applied. $\delta_a = 0^{\circ}$.



(b) Flap and aileron.

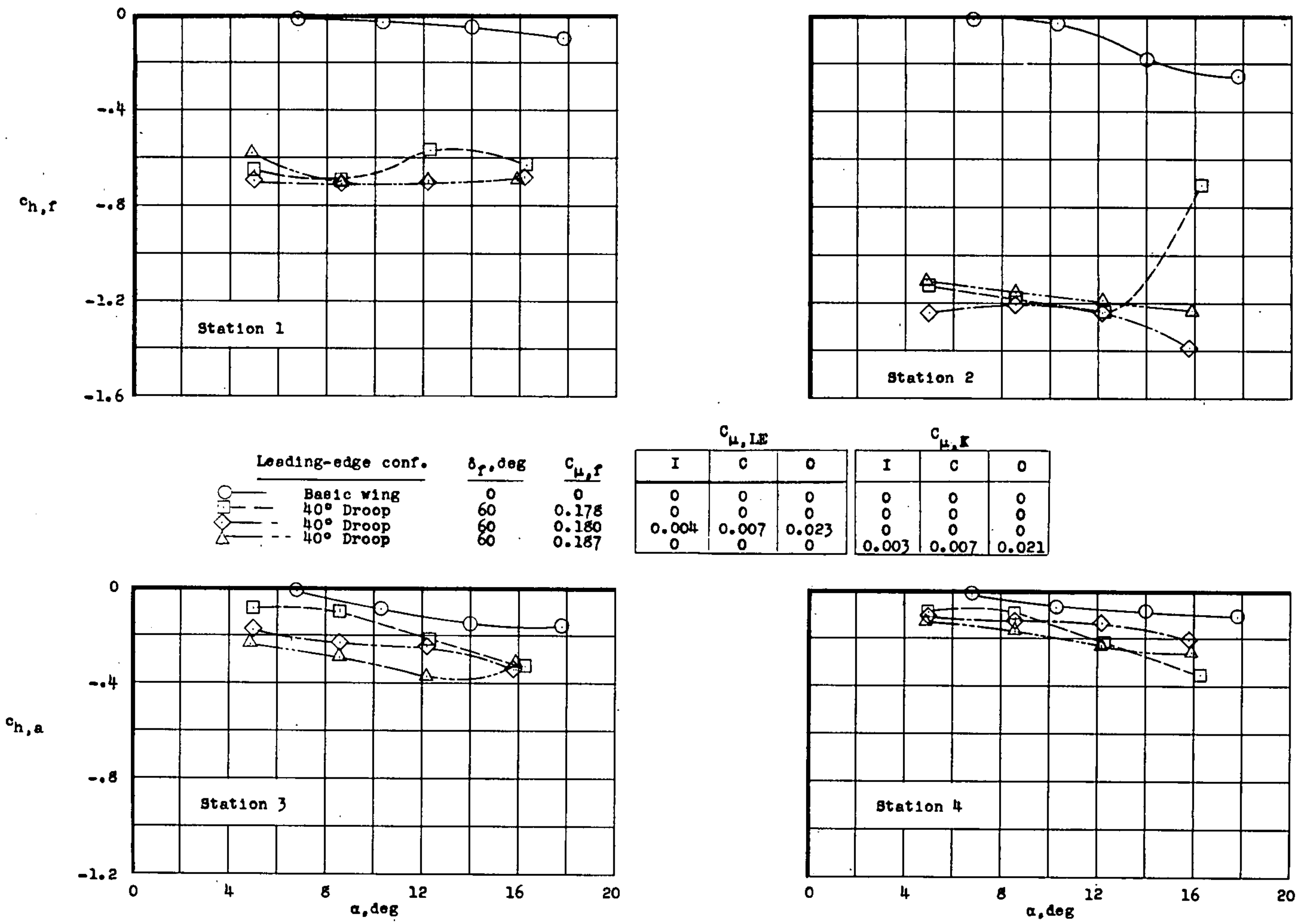
Figure 29.- Concluded.



Leading-edge conf.	δ_f,deg	$C_{\mu,f}$	$C_{\mu,LE}$ I	$C_{\mu,LE}$ C	$C_{\mu,LE}$ O	$C_{\mu,K}$ I	$C_{\mu,K}$ C	$C_{\mu,K}$ O
Basic wing	0	0	0	0	0	0	0	0
40° Droop	60	0.178	0	0	0	0	0	0
40° Droop	60	0.180	0.004	0.007	0.023	0	0	0
40° Droop	60	0.187	0	0	0	0.003	0.007	0.021

(a) Leading edge.

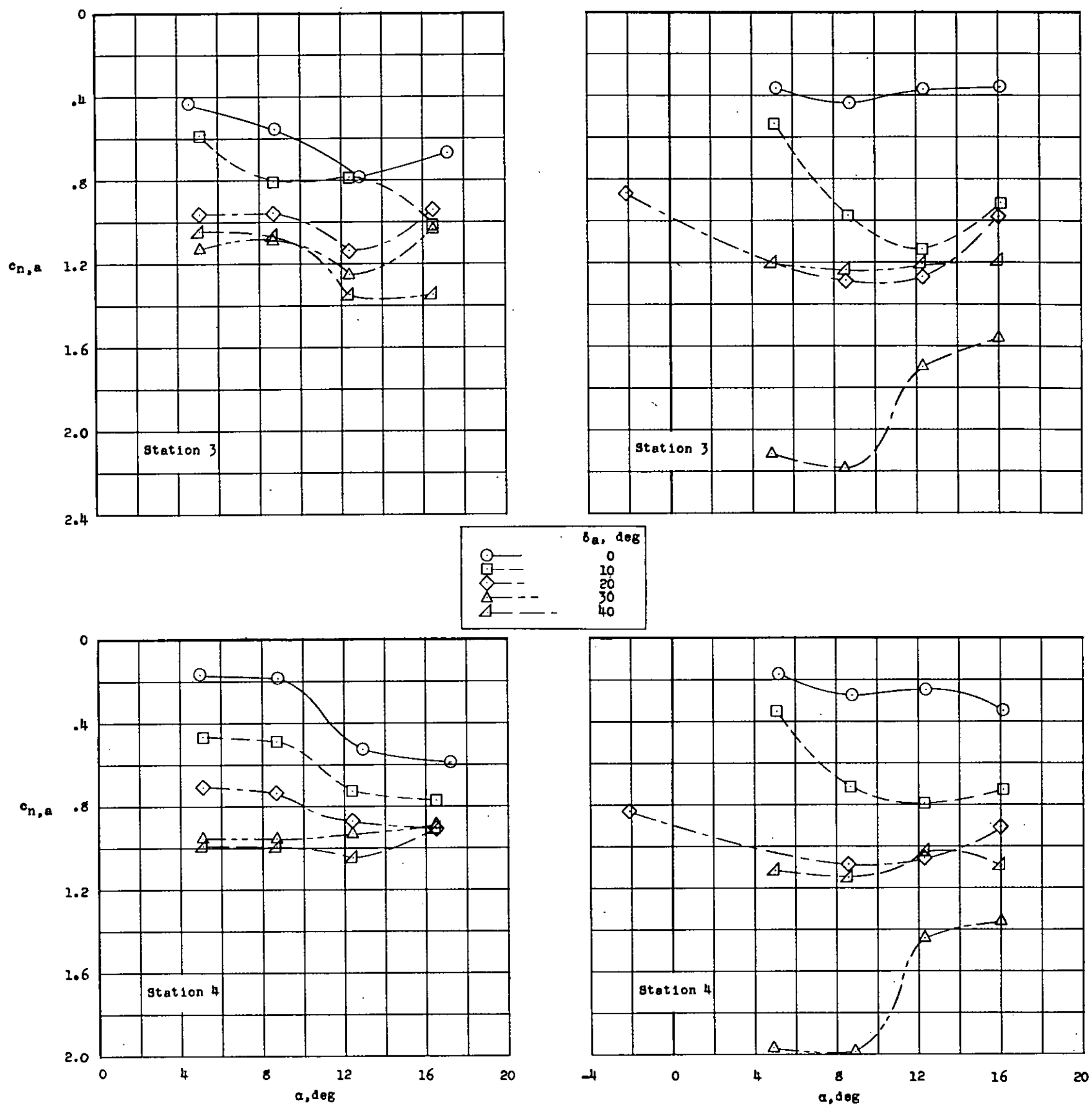
Figure 30.- Variation of leading-edge, flap, and aileron section hinge-moment coefficients with angle of attack, without and with leading-edge and flap blowing applied. $\delta_a = 0^{\circ}$.



Leading-edge conf.	δ_f,deg	$C_{\mu,f}$	$C_{\mu,LE}$ I	$C_{\mu,LE}$ C	$C_{\mu,LE}$ O	$C_{\mu,K}$ I	$C_{\mu,K}$ C	$C_{\mu,K}$ O
○ Basic wing	0	0	0	0	0	0	0	0
□ 40° Droop	60	0.178	0	0	0	0	0	0
◇ 40° Droop	60	0.180	0.004	0.007	0.023	0	0	0
△ 40° Droop	60	0.187	0	0	0	0.003	0.007	0.021

(b) Flap and aileron.

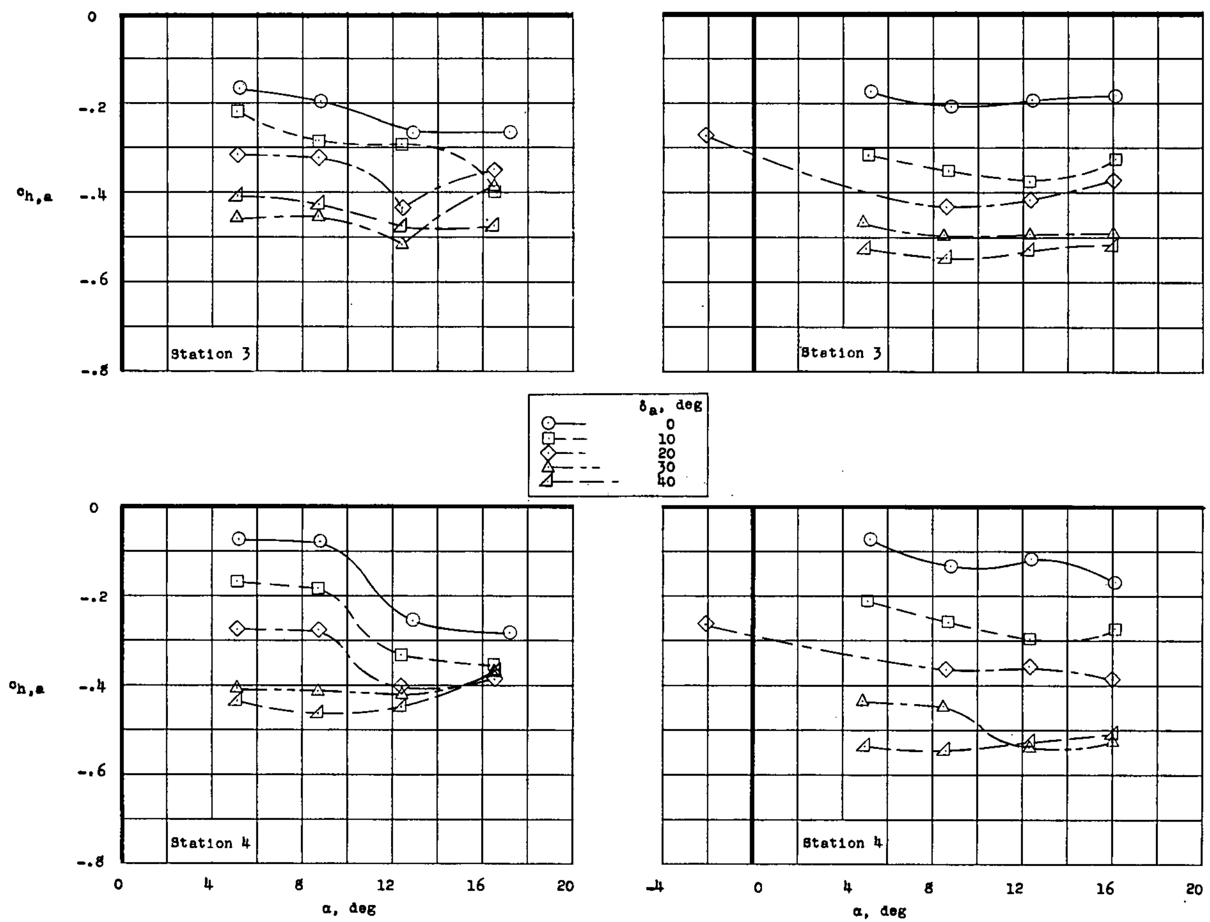
Figure 30.- Concluded.



(a) $C_{\mu,f} = 0.100$; $C_{\mu,K} = 0$.

(b) $C_{\mu,f} = 0.100$; $C_{\mu,K}$: I = 0, C = 0.008, O = 0.014.

Figure 31.- Variation of aileron section normal-force coefficient with angle of attack for several aileron deflections, with flap blowing alone and in combination with knee blowing, for the 40° drooped-leading-edge configuration. $\delta_f = 60^\circ$.



(a) $C_{\mu,f} \approx 0.100$; $C_{\mu,K} = 0$.

(b) $C_{\mu,f} \approx 0.100$; $C_{\mu,K}$: I = 0, C = 0.008, O = 0.014.

Figure 32.- Variation of aileron section hinge-moment coefficient with angle of attack for several aileron deflections, with flap blowing alone and in combination with knee blowing, for the 40° drooped-leading-edge configuration. $\delta_f = 60°$.