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REPORT No. 163

THE VERTICAL, LONGITUDINAL, AND LATERAL ACCEL- ERATIONS EXPERIENCED BY AN S. E. 5A AIR- PLANE WHILE MANEUVERING

By F. H. NORTON and T. CARROLL



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AERONAUTICAL SYMBOLS.

1. FUNDAMENTAL AND DERIVED UNITS.

	Symbol.	Metric.		English.	
		Unit.	Symbol.	Unit.	Symbol.
Length....	<i>l</i>	meter.....	m.	foot (or mile).....	ft. (or mi.).
Time.....	<i>t</i>	second.....	sec.	second (or hour).....	sec. (or hr.).
Force.....	<i>F</i>	weight of one kilogram.....	kg.	weight of one pound.....	lb.
Power....	<i>P</i>	kg.m/sec.....		horsepower.....	HP
Speed.....		m/sec.....	m. p. s.	mi/hr.....	M. P. H.

2. GENERAL SYMBOLS, ETC.

Weight, $W = mg$.

Standard acceleration of gravity,
 $g = 9.806\text{m/sec.}^2 = 32.172\text{ft/sec.}^2$

Mass, $m = \frac{W}{g}$

Density (mass per unit volume), ρ

Standard density of dry air, 0.1247 (kg.-m.-sec.) at 15.6°C. and 760 mm. = 0.00237 (lb.-ft.-sec.)

Specific weight of "standard" air, 1.223 kg/m.³ = 0.07635 lb/ft.³

Moment of inertia, mk^2 (indicate axis of the radius of gyration, k , by proper subscript).

Area, S ; wing area, S_w , etc.

Gap, G

Span, b ; chord length, c .

Aspect ratio = b/c

Distance from $c. g.$ to elevator hinge, f .

Coefficient of viscosity, μ .

3. AERODYNAMICAL SYMBOLS.

True airspeed, V

Dynamic (or impact) pressure, $q = \frac{1}{2} \rho V^2$

Lift, L ; absolute coefficient $C_L = \frac{L}{qS}$

Drag, D ; absolute coefficient $C_D = \frac{D}{qS}$

Cross-wind force, C ; absolute coefficient

$$C_c = \frac{C}{qS}$$

Resultant force, R

(Note that these coefficients are twice as large as the old coefficients L_c, D_c .)

Angle of setting of wings (relative to thrust line), i_w

Angle of stabilizer setting with reference to thrust line i_t

Dihedral angle, γ

Reynolds Number = $\rho \frac{Vl}{\mu}$, where l is a linear dimension.

e. g., for a model airfoil 3 in. chord, 100 mi/hr., normal pressure, 0°C: 255,000 and at 15.6°C, 230,000;

or for a model of 10 cm. chord, 40 m/sec., corresponding numbers are 299,000 and 270,000.

Center of pressure coefficient (ratio of distance of C. P. from leading edge to chord length), C_p .

Angle of stabilizer setting with reference to lower wing. $(i_t - i_w) = \beta$

Angle of attack, α

Angle of downwash, ϵ

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National Advisory Committee for Aeronautics

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SUMMARY.

This investigation was carried out by the Langley Field Laboratory of the National Advisory Committee for Aeronautics for the purpose of measuring the accelerations along the three principal axes of an airplane while it was maneuvering. The airplane selected for this purpose was the fairly maneuverable S. E. 5 A. and the instruments used were the N. A. C. A. three component accelerometer and the N. A. C. A. recording airspeed meter. The results showed that the normal accelerations did not exceed 4.00 *g*. while the lateral and longitudinal accelerations did not exceed 0.60 *g*.

INTRODUCTION.

The National Advisory Committee for Aeronautics has conducted in the past several investigations on the forces normal to the wings experienced by an airplane in maneuvering. The tests were made however on airplanes of the training type, so that it was felt desirable, now that the committee had a combat airplane in good condition, to determine the loadings on this type of airplane during maneuvers. It was also thought that a record of the accelerations along the longitudinal and lateral axes would also be of interest because as far as it is known no such records have previously been taken. The N. A. C. A. three component accelerometer which has been recently developed allowed the recording of the three accelerations simultaneously.

The maneuvers carried out were the usual ones of a loop, roll, spin, a right and left wing-over turn, someside-slips and some skids. It should be kept in mind that the loads experienced are by no means as great as could be obtained by very rough handling.

Below are given the references to the more important investigations on accelerations in flight:

Accelerometer Design.—N. A. C. A. Report No. 100. 1921.

Accelerations in Flight.—N. A. C. A. Report No. 99. 1921.

A Study of Airplane Maneuvers with Special Reference to Angular Velocities.—N. A. C. A. Report No. 155. 1922.

The N. A. C. A. Three Component Accelerometer.—N. A. C. A. Technical Note No. 112. 1922.

Preliminary Report on the Measurement of Accelerations on Aeroplanes in Flight.—R. & M. No. 376. 1917.

Report on the Measurement of Accelerations on Aeroplanes in Flight.—R. & M. No. 469. 1918.

Notes on the Design of a Recording Three Dimensional Accelerometer for Use in Aeroplanes.—R. & M. No. 627. 1919.

Development of an Airplane Shock Recorder.—A. F. Zahm. Journal of the Franklin Institute, August, 1919.

APPARATUS AND METHOD.

The airplane used in this investigation was a standard S. E. 5A. with a Wright model E engine, but with no military load (fig. 1). The accelerometer was the N. A. C. A. three-component instrument which has been previously described. This instrument was carefully mounted on sponge rubber within a few inches of the center of gravity of the machine, in order that it should not be affected by angular accelerations. The recording airspeed meter was connected to a swivelling pitot head on the wing strut and a calibration was made by flying this airplane alongside of another airplane whose airspeed meter had been carefully calibrated. Samples of the records obtained are shown in figure 2.

PRECISION.

The airspeed records have a precision of better than ± 2 miles per hour, and the values of the accelerations are accurate to $\pm 0.05 g$.

RESULTS.

The airspeed and accelerations for the various maneuvers are plotted in figures 3 to 7 on a uniform time base. In all cases the acceleration is considered positive when the air force, along its corresponding axis, acts in a positive direction, and the longitudinal axis was taken parallel to the top longeron. All of the accelerations given are the total accelerations acting on the airplane; that is, the component of gravity is included. The maximum accelerations experienced along each axis during the various maneuvers are summarized in a table below:

Maneuver.	X	Y	Z
Loop.....	-0.50 g.	-0.10 g.	3.80 g.
Spin.....	-.20	$\pm .35$	3.20
Roll.....	-.50	$\pm .14$	2.92
Wing over.....	-.60	$\pm .20$	2.60
Skid.....	.00	$\pm .23$	1.20
Slip.....	-.15	$\pm .40$	1.20

The greatest acceleration along the Z axis occurred in a loop and amounted to 3.80 g.; along the Y axis the greatest acceleration occurred in a side-slip and was $-0.40 g$.; and the maximum acceleration of 0.60 g. along the X axis occurred in a wing-over.

It is interesting to compare the normal acceleration on this airplane with those found on the JN4h in Report No. 99. The S. E. 5A. showed a slightly greater loading in the loop, the same acceleration in a spin, and a markedly lower acceleration in a roll. It is noticeable that the S. E. 5A rolls smoothly and easily, whereas the JN4h must be forced through a roll with a very high initial speed.

CONCLUSIONS.

It may be concluded from these tests that the accelerations acting along the longitudinal and lateral axes are very small compared with the acceleration along the vertical axis. It is also shown that the normal acceleration experienced by an airplane such as the S. E. 5A. in ordinary maneuvering are no higher than for a training airplane of the JN4h type.

It should be noted that the accelerations in a given maneuver and with a given airplane are dependent on the manner in which the pilot handles the controls. If he is rough the airplane will be heavily loaded, but if he is skilful the loadings will be small. In general, however, the pilot feels the accelerations and unconsciously keeps from reaching high values, except under the stress of a combat when the loads may be much higher than in ordinary maneuvering.

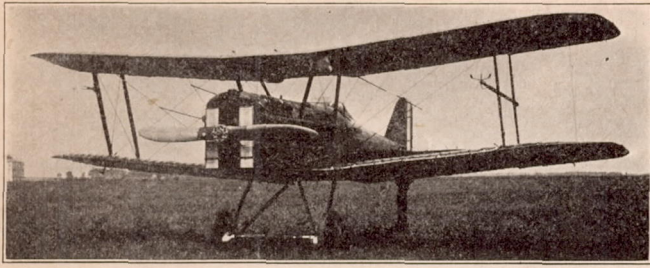
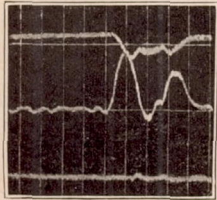
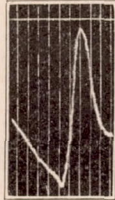


FIG. 1.—S. E. 5A, used in these tests.



Accelerations.



Air speed.

FIG. 2.

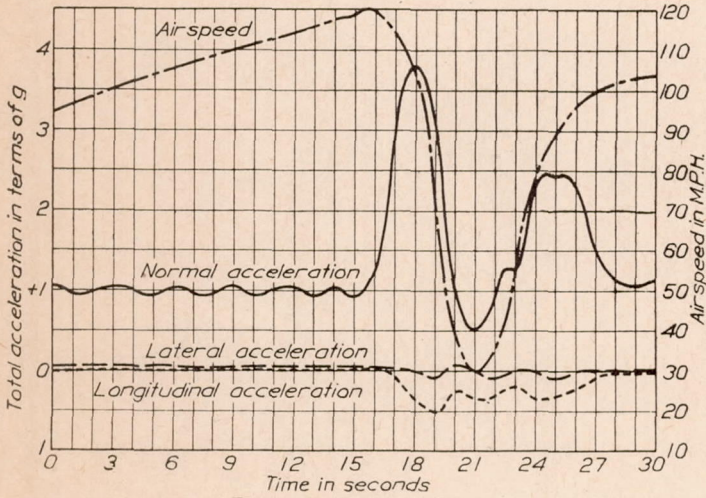


FIG. 3.—Accelerations in loop.

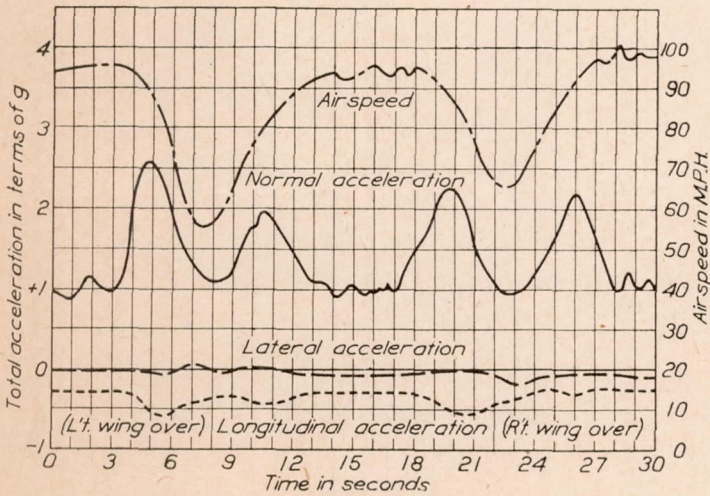


FIG. 4.—Accelerations in wing over.

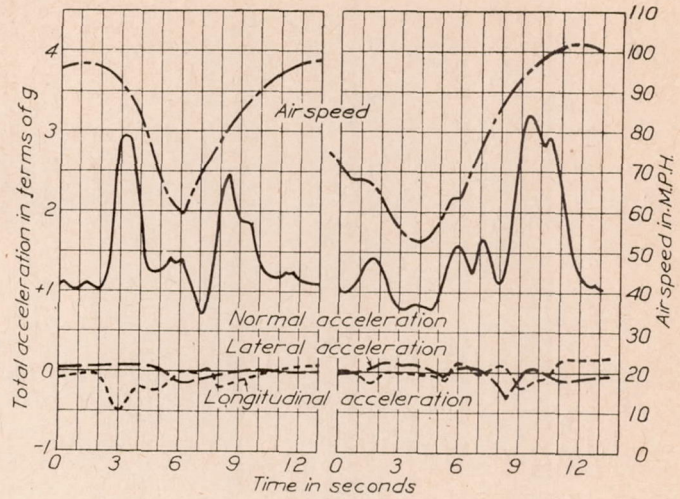


FIG. 5.

Accelerations in a R. roll.

Accelerations in a R. spin, two turns

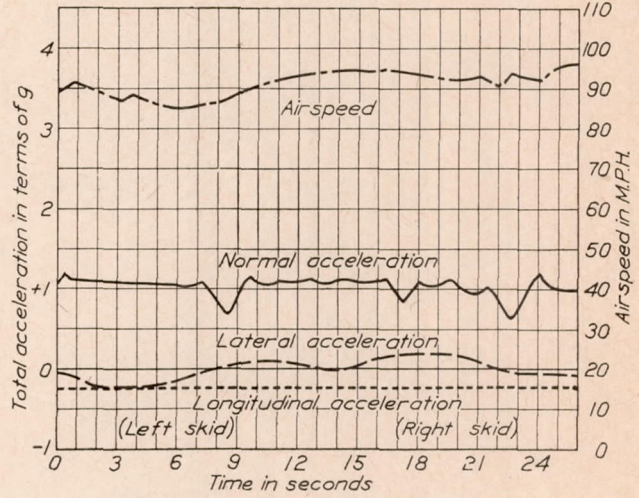


FIG. 6.—Accelerations in skid.

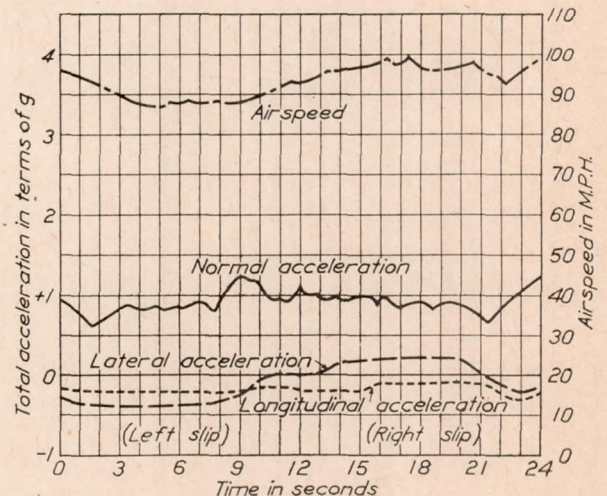
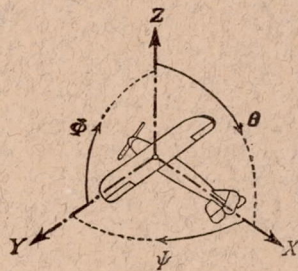


FIG. 7.—Accelerations in slip.



Positive directions of axes and angles (forces and moments) are shown by arrows.

Axis.		Force (parallel to axis) symbol.	Moment about axis.			Angle.		Velocities.	
Designation.	Sym- bol.		Designa- tion.	Sym- bol.	Positive direc- tion.	Designa- tion.	Sym- bol.	Linear (compo- nent along axis).	Angular.
Longitudinal.....	X	X	rolling.....	L	Y → Z	roll.....	Φ	u	p
Lateral.....	Y	Y	pitching....	M	Z → X	pitch.....	Θ	v	q
Normal.....	Z	Z	yawing.....	N	X → Y	yaw.....	Ψ	w	r

Absolute coefficients of moment

$$C_l = \frac{L}{q b S} \quad C_m = \frac{M}{q c S} \quad C_n = \frac{N}{q f S}$$

Angle of set of control surface (relative to neutral position), δ . (Indicate surface by proper subscript.)

4. PROPELLER SYMBOLS.

Diameter, D
 Pitch (a) Aerodynamic pitch, p_a
 (b) Effective pitch, p_e
 (c) Mean geometric pitch, p_g
 (d) Virtual pitch, p_v
 (e) Standard pitch, p_s
 Pitch ratio, p/D
 Inflow velocity, V'
 Slipstream velocity, V_s

Thrust, T
 Torque, Q
 Power, P
 (If "coefficients" are introduced all units used must be consistent.)
 Efficiency $\eta = T V/P$
 Revolutions per sec., n ; per min., N
 Effective helix angle $\Phi = \tan^{-1} \left(\frac{V}{2\pi r n} \right)$

5. NUMERICAL RELATIONS.

1 HP = 76.04 kg. m/sec. = 550 lb. ft/sec.
 1 kg. m/sec. = 0.01315 HP
 1 mi/hr. = 0.44704 m/sec.
 1 m/sec. = 2.23693 mi/hr.

1 lb. = 0.45359 kg.
 1 kg. = 2.20462 lb.
 1 mi. = 1609.35 m. = 5280 ft.
 1 m. = 3.28083 ft.