

NASA Technical Memorandum 106826
AIAA-95-0538

11
3
p-13

Ice Accretion with Varying Surface Tension

Alan J. Bilanin
Continuum Dynamics, Inc.
Princeton, New Jersey
for
NYMA, Inc.
Engineering Services Division
Brook Park, Ohio

David N. Anderson
Lewis Research Center
Cleveland, Ohio

Prepared for the
33rd Aerospace Sciences Meeting and Exhibit
sponsored by the American Institute of Aeronautics and Astronautics
Reno, Nevada, January 9-12, 1995

(NASA-TM-106826) ICE ACCRETION
WITH VARYING SURFACE TENSION
(NASA. Lewis Research Center) 13 p

N95-19285

Unclass



National Aeronautics and
Space Administration

G3/03 0035009



Ice Accretion with Varying Surface Tension

Alan J. Bilanin
Continuum Dynamics, Inc.
and

David N. Anderson
NASA Lewis Research Center
Cleveland, OH

Abstract

During an icing encounter of an aircraft in flight, super-cooled water droplets impinging on an airfoil may splash before freezing. This paper reports tests performed to determine if this effect is significant and uses the results to develop an improved scaling method for use in icing test facilities. Simple laboratory tests showed that drops splash on impact at the Reynolds and Weber numbers typical of icing encounters. Further confirmation of droplet splash came from icing tests performed in the NASA Lewis Icing Research Tunnel (IRT) with a surfactant added to the spray water to reduce the surface tension. The resulting ice shapes were significantly different from those formed when no surfactant was added to the water. These results suggested that the droplet Weber number must be kept constant to properly scale icing test conditions. Finally, the paper presents a Weber-number-based scaling method and reports results from scaling tests in the IRT in which model size was reduced up to a factor of 3. Scale and reference ice shapes are shown which confirm the effectiveness of this new scaling method.

θ	Air-energy transfer term in energy equation, K
Λ_f	Latent heat of freezing, cal/gm
Λ_v	Latent heat of vaporization, cal/gm
ρ	Density, dyne/cm ³
μ	Viscosity, gm/cm s
$\sigma_{w/a}$	Surface tension of water against air, dyne/cm
τ	Icing time, min

Subscripts:

a	Air
δ	Based on droplet size
f	freezing point
i	Ice
m	Measured
R	Reference size and conditions
$surf$	Surface
S	Scale size and conditions
tot	Total
T	Theoretical
w	Water

Nomenclature

A_c	Accumulation parameter, dimensionless
b	Relative heat factor, dimensionless
c	Characteristic model length, cm
c_p	Specific heat, cal/gm K
h_c	Convective film heat-transfer coefficient, cal/sec m ² K
k	Thermal conductivity, cal/sec m K
LWC	Liquid-water content, g/m ³
M	Mach number, dimensionless
n	Freezing fraction, dimensionless
Nu	Nusselt number, dimensionless
p	Ambient static pressure, nt/m ²
p_w	vapor pressure of water, nt/m ²
R_a	Gas constant for air, 287.0 nt m/kg K
Re	Reynolds number, dimensionless
t	Ambient static temperature, °C
T	Ambient static temperature, K
U	Droplet velocity, m/s
V	Airspeed, m/s
We	Weber number, dimensionless
δ	Droplet median volume diameter, μ m
Δ	Film thickness or ice thickness, m
ϕ	Droplet-energy transfer term in energy equation, K

Introduction

Wind tunnel icing tests of subscale aerodynamic components have been undertaken for decades. Yet, scaling laws which relate tunnel data to predict what would be anticipated during operation of the component in the atmosphere have yet to be agreed upon. This is evident from reviews of scaling laws by Bilanin¹ in 1988 and Anderson² in 1994. There exist about half a dozen methodologies which have competed over the years for international acceptance, but in spite of years of validation testing, to date no methodology has shown a clear advantage. One reason is that until quite recently, the ability of a wind tunnel test engineer to set icing test conditions such as droplet size and liquid-water content, has been hindered with instrumentation and calibration shortcomings. The second, and probably more important, reason is that almost without exception previous scaling methodologies have ignored water droplet impact and film dynamics in the scaling analysis. This observation was brought to the attention of the icing community in 1988¹. This paper reports on the first attempt to provide experimental evidence that droplet impact and water film dynamics cannot be neglected in the derivation of scaling laws, especially under icing conditions where low freezing fractions are anticipated. The results presented here are significant not just for scaling but they also suggest that ice accretion codes need to implement droplet impact and liquid-film dynamic models to

improve the accuracy of predictions. Discrepancies between predicted and measured ice shapes are often blamed on inaccuracies in modeling the convective heat transfer coefficient. We suspect that some of this discrepancy could be explained by considering droplet-surface interactions.

Evidence that all is not well with past scaling methodologies was presented in reference 1 where it was argued that if both the accumulation factor, A_c , and the freezing fraction, n , were accurately known at each point along an aerodynamic component, the prediction of ice accretion and a comparison between this prediction and test data must agree favorably. It was then shown using test data along with the scaling methodology in the SIMICE scaling code³ that seriously misunderstood discrepancies exist. The SIMICE code computes the freezing fraction according to the Messinger analysis⁴. Figure 1 is taken from reference 1 and shows the measured and predicted ice thickness at the stagnation point for two tests having predicted freezing fractions, n_p , of .15 and .5. The stagnation-point ice thickness is exactly proportional to $A_c n$, and the predictions in figure 1 were based on this. The actual measured freezing fractions, n_m , were .35 and .6, respectively. Ice thicknesses for the lower freezing fraction were greater than predicted by a factor of 2.3. Thus, either the freezing fraction computation or the estimate of accumulation parameter were in error.

The suspicion that something may not be correct with the methodology for computation of the collection factor is motivated by the observation of raindrops impacting a shallow puddle. The schematic of this collision is shown in figure 2. Here, for the sake of argument and to good approximation during the interaction of the droplet with the film, we will neglect the shear forces of the air on the droplet. Then the droplet film dynamics are controlled by only three nondimensional parameters:

Δ/δ	film thickness/droplet diameter
$\rho_w U \delta / \mu_w$	Reynolds number
$\rho_w U^2 \delta / \sigma_{w/a}$	Weber number

Also, we know from observations that rain drops splash upon impacting a water layer. In Table I, we have computed and tabulated the typical order of magnitude values of these three nondimensional parameters for both rain and icing conditions. Quite surprising is the result that in general the values of the Reynolds and Weber numbers for rain are typical of the values during icing. Hence, one can study the impact of 1-mm raindrops moving at 10 m/s impacting a 1-mm film instead of studying the impact of a 20- μ m droplet impacting a film at 500 kts.

This paper presents the results of simple experiments to confirm the importance of droplet splashing at Re and We representative of icing encounters. Icing tests in the NASA Lewis Icing Research Tunnel are described in which the surface tension of the spray was altered by the addition of a surfactant. Finally, a practical scaling method is described and tested in which the droplet We is maintained the same between scale and reference conditions.

Droplet Splashing Tests

The above discussion mentioned the use of slow-moving large water droplets impacting a water film to determine if film dynamics and splashing can be neglected in the development of scaling laws and in computer models of the ice accretion. In this section, we report on simple tests which have been conducted to examine droplet impact dynamics relevant to icing.

Figure 3 is a sketch of the droplet-impact rig used during testing. A burette is mounted at distance h ($h < 5$ m) above a small dish sitting on a milligram scale. By slowly allowing the water to drop from the burette and counting the number of drops produced per milliliter, one can easily compute a typical droplet diameter. From previous work involving spray dispersion in aircraft vortex wakes⁵ for water droplets released from rest into still air, the velocity of the droplet as a function of release height, h , is known. In the tests reported here, the droplet diameter was 4.7 mm and the impact velocity from a release height of 1 and 4.7 m was 4.2 and 8.1 m/s, respectively.

The results are plotted on figure 4 where drops impact a film which is 1/3, 2/3, 3/4 and 4/3 of the impacting drop diameter. Plotted on the vertical axis is the percent of mass gained after the impact. For example, if the incoming drop's mass is completely added to the film, the percent mass gain would be 100%. Even at the lower impact velocities of 4.2 m/s, on the order of 25% of the incoming mass is splashed away. At the highest impact velocities, a more surprising result is obtained in that it is possible to lose mass of liquid from the film. A mass loss of 100% implies that for every incoming droplet, two droplets are splashed away. Note that the results are strongly dependent on the film-thickness-to-droplet-diameter ratio, and film thicknesses which are less than the diameter of the incoming drop seem to experience the most mass ejection from the film. Note that each test condition was repeated a minimum of three times, and the repeatability appears to be quite high.

Figure 4 gives the Weber and Reynolds numbers for the tests conducted. These can be compared to Table I where it can be seen that these values are in fact typical of those anticipated during icing. The above results therefore strongly suggest that droplet splashing and droplet impact dynamics with a liquid film may be important in the development of scaling laws and also must be modeled if ice accretion computer codes are to make physically realistic predictions of ice accretion.

Icing Tunnel Experiments

If droplet splashing has an effect on the ice-accretion process, then varying the surface tension of the spray water will change the ice shape. Two tests were performed in the NASA Lewis Icing Research Tunnel (IRT) to evaluate the effect of surface tension on ice shape. For the first test, Liqui-Nox, a commercial cleaning solution, was added to the spray-bar water supply so that the entire IRT spray had reduced surface tension. For the second test, Kodak Photo-Flo 600 was added to the spray of a single nozzle directed at the center of the test section while the IRT spray bars maintained

their normal spray of demineralized water.

The results of these tests led to the development of a scaling method based on the requirement that the Weber number, We , be matched between scale and reference tests. Additional tests were performed in the IRT to evaluate this method. The surfactant-addition tests, the constant- We scaling method and the scaling tests and results will be described in this section.

NASA Lewis Icing Research Tunnel. The IRT is shown in figure 5. It has been described in reference 6. The IRT has a test section width of 2.74 m (9 ft) and a height of 1.83 m (6 ft.) It is capable of operation at test-section airspeeds up to 160 m/s (350 mph.) A refrigeration system permits accurate control of the test-section temperature from 230 to 278 K (-40 to 40°F.) A water-spray system⁷ with 8 spray bars provides the ability to control test-section liquid-water content from .2 to 3 g/m³ and droplet median volume diameters from 15 to 40 μ m.

Two sets of spray nozzles, known as the mod-1 and standard nozzles, are used in the IRT to provide different ranges of liquid-water content and droplet size⁷. For these tests, only the mod-1 nozzles were used.

For tests using a single nozzle to add surfactant, a mod-1 nozzle was mounted on the spray-bar support and aimed so that its spray struck the model at the centerline of the test section. This nozzle was supplied with air from the spray-bar manifold while it received water from a tank independent of the spray-bar supply. The single-nozzle water pressure was controlled to the same value as the water pressure for the spray bars. When the single nozzle was used, surfactant was added to its water supply while the main spray system used demineralized water with no additive. It was found by ice shape comparisons when demineralized water with no surfactant was sprayed from the single nozzle, that of the total water reaching the center of the model, about 10% came from the single nozzle and the rest from the main spray bars. Consequently, for all the single-nozzle tests the spray-bar pressures were set for a liquid-water content of 90% the desired value.

The tests of the constant- We scaling method were performed without surfactant.

Test Hardware. Ice accretion was measured on hollow circular aluminum cylinders. Each cylinder was mounted vertically in the center of the test section. Cylinders with 15.2-, 7.6-, 5.1- and 2.5-cm (6-, 3-, 2- and 1-in) diameters were used. Figure 6 shows how each cylinder was positioned in the IRT test section. A retractable shield was positioned in front of the cylinder to protect it from spray during the spray-bar start-up period when the water and air pressures were stabilizing. The shield could be retracted rapidly into the tunnel ceiling by remote operation of a hydraulic actuator.

Test Procedure. Tests were performed by first establishing the desired velocity and temperature. Water spray conditions were then selected, and when tunnel conditions had stabilized, the water spray was initiated. The shroud protecting the test cylinder from the spray was lifted when the spray conditions had stabilized, and

the spray timer was started at this time. When the prescribed spray period was completed, the spray was shut off and the tunnel brought to idle to permit personnel entry into the test section. The ice shape was recorded by first melting a thin slice through the ice normal to the cylinder axis. The shape was traced onto a cardboard template; these shapes were later digitized and recorded on a computer disk. After the ice shape was recorded, the model was cleaned and the procedure repeated for the next spray condition.

Results: Surface-Tension Experiments

Surface Tension Measurements The surface tension of the demineralized IRT water, two dilutions of water:Liqui-Nox mixtures, two dilutions of water:Photo-Flo 200 mixtures and a water:Photo-Flo 600 mixture was measured. Photo-Flo 600 is a more concentrated form of Photo-Flo 200 so that a 600:1 mix of water to Photo-Flo 600 is equivalent to a 200:1 mix of water to Photo-Flo 200. Measurements were made on three occasions using an apparatus which employed the ring-detachment method, and the results are given in Table I. The published value⁸ of surface tension for water at 20°C is 73 dyne/cm. When there are impurities present, however, the surface tension will be less than this. It is not surprising, then, that the surface tension of water was found to be lower than the published value. The addition of surfactant in the form of Liqui-Nox or either of the Photo-Flo concentrations reduced the surface tension of the IRT water to about half its untreated value. Furthermore, increasing the dilution of the water:surfactant mixture had virtually no effect on the surface tension.

Effect of Reduced Surface Tension on Ice Shape Initial tests with surfactant addition were made with Liqui-Nox added to the entire spray bar water supply. At the conclusion of these tests, it was necessary to flush the entire spray bar system thoroughly to remove any traces of the surfactant for subsequent icing tests with demineralized water. To avoid the inconvenience of this purging procedure, a single spray nozzle with its own water supply was mounted on the spray-bar array in a such a position that its spray would strike the model at the center of the test section. Thus, the single nozzle could be operated with surfactant-treated water while the full spray-bar system operated with untreated demineralized water. Because ice shapes were only measured at the center of the model for this study, this arrangement would be effective if it produced the same ice shapes as resulted from adding surfactant to the full spray-bar system.

To test the single-nozzle system, Kodak Photo-Flo 600 in a mixture of 600:1 water:Photo-Flo was used in the single-nozzle water supply. A comparison of centerline ice shapes using the single-nozzle and full-spray-bar surfactant addition methods is shown in figure 7. The solid line represents the shape resulting from adding Liqui-Nox to the spray-bar water supply. The dotted line indicates the ice shape accreted when the single nozzle operated with the water:Photo-Flo mixture. The shapes of the ice formed by either method of delivering surfactant agree within the normal repeatability of the tunnel. This result verifies that the simpler single-nozzle method of surfactant addition gives results at the centerline which are equivalent to the less-desirable method of

adding surfactant to the full spray-bar system.

As noted above, the single nozzle contributed only about 10% of the total water reaching the model centerline. Thus, most of the incoming drops would not have the reduced surface tension when the single nozzle was used to add surfactant. Apparently, then, the impacting droplet surface tension was of less importance to the physics of the process than the surface tension of the unfrozen water on the surface of the model. Because dilution has little effect on surface tension, once the treated droplets mixed with water on the surface, the surface tension of the surface mixture would be expected to be about the same for either method of adding surfactant.

To test the effect of surfactant on ice shape, conditions were run which had previously been used with unadulterated demineralized water. The results are shown in figure 8. The solid line gives the shape at the centerline produced with demineralized water, and the dotted line represents the shape when surfactant was added to the spray. In figure 8(a) results are given for tests in which 1 part of Liqui-Nox cleaner was added to 400 parts (by volume) of demineralized water. The addition of surfactant totally changed the shape of the horns and reduced the horn angle.

Results for the single-nozzle surfactant-addition tests are given in figure 8(b). Again, surfactant addition had a dramatic effect on the shape of the ice. Although droplet splashing has been shown in the bench-top experiments described previously in this paper to be significant, an additional effect of the surfactant on ice shape may also result from a reduction in droplet size. When the surface tension is halved, the droplet size can be expected to decrease by about 25%⁹. For the single-nozzle tests only about 10% of the droplets have reduced surface tension, however, so the median volume diameter of the total spray should be close to that for water without surfactant. The freezing temperature of the water:Photo-Flo mixture was tested and found to be the same as water. Although the mixture latent heat was not measured for this study, it should be the same as water because of the extremely small amount of surfactant used. Thus, the primary reason for the differences in ice shape with and without surfactant is apparently the change in the surface tension of unfrozen water on the surface of the model. This observation reinforces the conclusion of the droplet-splashing tests that droplet-surface interaction plays a significant role in the ice-accretion process.

Implications for Scaling

The strong effect of droplet-surface interaction demonstrated by both the splashing tests and the icing-tunnel tests suggests that the Weber number needs to be considered in developing scaling methods. Rigorous scaling would require that the Reynolds and Weber numbers be matched between scale and reference conditions¹. To do this it is necessary that, approximately,

$$V_S = V_R \frac{c_R}{c_S} \quad (1)$$

and

$$\sigma_S = \sigma_R \frac{c_R}{c_S} \quad (2)$$

Thus, for a half-size model, for example, both the airspeed and the surface tension of the spray need to be roughly doubled over their desired reference values. While it is relatively easy to *reduce* surface tension by adding surfactant to the spray water, *increasing* the surface tension of the spray to satisfy equation (2) does not appear to be practical. Furthermore, the airspeeds required by equation (1) for small-scale models will generally produce unacceptably high Mach numbers. It seems, then, that this approach to scaling is likely to be unusable for most scaling situations. However, a practical scaling method can be developed by relaxing the need to match Re between scale and reference situations. This is a reasonable simplification, for the Re should have little effect on the flowfield, provided the flow becomes turbulent over some part of the body. With glaze ice shapes, the point of separation and transition are determined by the ice shape rather than by the Re . Consequently, a scaling method was developed which is based on maintaining the same We for both scale and reference conditions and ignoring the Re . This method is discussed in the following sections.

Constant-Weber-Number Scaling. The method is outlined here. Results from preliminary tests to verify the method in the IRT are described in the next section.

Unlike other published scaling methods², this method permits only the choice of scale model size, c_S . The reference model size, c_R and all reference conditions are, of course, known. The following four equations are first solved simultaneously to give the scale airspeed, V_S , scale static temperature, T_S , scale static pressure, p_S , and scale drop size, δ_S :

$$V_S = V_R \left(\frac{\delta_R}{\delta_S} \right)^{\frac{1}{2}} \quad (3)$$

$$T_S = T_R + \frac{V_R^2}{2 c_{p,w,ref}} - \frac{V_S^2}{2 c_{p,w,ref}} \quad (4)$$

$$p_S = \frac{P_{tot,S}}{1 + \frac{V_S}{2 R_a T_S}} \quad (5)$$

$$\delta_S = \delta_R \left(\frac{c_S}{c_R} \right)^{.62} \left(\frac{p_S}{p_R} \right)^{.24} \left(\frac{V_S}{V_R} \right)^{-38} \quad (6)$$

Equation (3) results from matching the scale and reference We , and equation (4) is obtained by matching droplet energy transfer terms (see equation (9), below) in the Messinger energy balance⁴ at the model surface. There is no fundamental necessity that the droplet energy terms be matched; however, an equation is needed to solve for the static temperature, and equation (4) is convenient. Equation (5) simply relates the static pressure in the tunnel to the known ambient pressure (tunnel total pressure) and equation (6) results from matching the droplet trajectories.

The freezing fraction, n , was defined by Messinger⁴ as the fraction of impinging water which freezes in the impingement zone. The scale LWC is found by equating the scale and reference freezing fraction. From the Messinger energy equation, the freezing fraction is

$$n = \frac{c_{p,w}}{\Lambda_f} \left(\phi + \theta \frac{h_c}{LWC V \beta_0 c_{p,w}} \right) \quad (7)$$

Thus,

$$LWC_s = \frac{\theta_s h_{c,s}}{V_s \beta_0 c_{p,w} \left(\frac{\theta_R h_{c,R}}{LWC_R V_R \beta_0 c_{p,w}} + \phi_R - \phi_s \right)} \quad (8)$$

In equations (7) and (8), ϕ is the droplet energy transfer term,

$$\phi = T_f - T - \frac{V^2}{2c_{p,w}} \quad (9)$$

It was noted above in reference to equation (4) that ϕ_s and ϕ_R are equated in this scaling method. Furthermore, the collection efficiency, β_0 , must be the same for scale and reference tests. Thus, equation (8) simplifies to

$$LWC_s = LWC_R \frac{\theta_s h_{c,s} V_R}{\theta_R h_{c,R} V_s} \quad (10)$$

From the Messinger equation, θ is the air energy transfer term:

$$\theta = T_{surf} - T - r \frac{V^2}{2c_{p,a}} + .693 \frac{\text{gm K}}{\text{joule}} \Lambda_v \frac{P_{w,surf} - P_w}{p} \quad (11)$$

where r is the recovery factor, taken as .875 in this analysis, $p_{w,surf}$

is the vapor pressure of water at the surface of the model (i.e., at T_{surf}) and p_w is the vapor pressure of water in the atmosphere (i.e., at T). The vapor pressures for this study were from Pruppacher and Klett⁸.

The scale and reference convective heat transfer coefficients, $h_{c,s}$ and $h_{c,R}$ in equation (10) were evaluated using the following empirical expression from Gelder and Lewis¹⁰:

$$h_c = 1.244 \frac{k_a}{c} \left(\frac{V c \rho_a}{\mu_a} \right)^{.4774} \quad (12)$$

The final scale parameter needed is the spray time, τ . It was found by matching the scale and reference accumulation parameters:

$$\frac{LWC_s V_s \tau_s}{\rho_1 c_s} = \frac{LWC_R V_R \tau_R}{\rho_1 c_R} \quad (13)$$

Table III shows results of calculations applying this scaling method to five reference conditions. In reference 2 a number of published scaling methods were shown to work successfully with rime ice but not for glaze. For this reason, reference conditions were chosen to test this new method with the more-difficult-to-scale glaze ice.

Verification of Constant-Weber-Number Scaling Method. The proposed scaling method was applied in a series of tests in the IRT using the ice-accretion test techniques discussed earlier. Scale sizes of 1/2 and 1/3 the reference cylinder diameters were used.

Due to a transducer calibration error, both reference and scale droplet size, δ and liquid-water content, LWC , were found after the completion of the test to differ from those planned. Table III gives the reference conditions (first line of each test case) which resulted when the test conditions were corrected to give the true droplet size and LWC for the five test cases. The scale conditions listed in Table III then were obtained by applying the constant- We scaling method to these corrected reference conditions. The actual scale conditions tested are given in Table IV. Comparing the scale conditions in Table IV with those in Table III, one can see that the scale airspeeds tested were as much as 8% lower and scale droplet sizes were up to 18% lower than the values required by this scaling method. Other scale test conditions were close to those required. The scale Weber numbers tested were lower than the reference values for these tests by as much as 28%.

Figure 9 shows the ice shapes for the conditions of case A in Table IV. The solid line represents the ice shape which resulted from testing at the reference conditions. The dashed line is the scale result. The scale ice shape coordinates have been multiplied by the inverse of the scale factor so that the shapes can be compared directly. The scale test for case A was performed on a 7.6-cm cylinder and resulted in horn-glaze ice very similar in overall shape and quantity of ice to the reference conditions tested on a 15.2-cm

cylinder. Figures 10 and 11 are also for reference tests with 15.2-cm cylinders but with conditions giving different forms of glaze ice. In each case the scale test, using a 7.6-cm cylinder, gave shapes very close to the reference shapes.

Figures 12 and 13 present results for testing with a 7.6-cm reference cylinder and 5.1-cm and 2.5-cm scale cylinders. The results for the 5.1-cm cylinders are shown in dashed lines and the ice shapes for the 2.5-cm cylinders are represented with dotted lines. Figure 12 shows results for horn glaze ice, and figure 13 is for glaze ice at a warmer temperature. In each case, the two scale tests faithfully reproduced the reference ice shape.

Agreement between scale and reference ice shapes was significantly better using the constant- We scaling method than has been demonstrated in past studies using other scaling methods in the IRT².

Concluding Remarks

This study has demonstrated the importance of droplet splash to the ice-accretion process. Simple tests performed with Re and We of magnitudes typical of those in icing encounters showed that droplet splash can have a significant effect on surface dynamics. Tests were also performed in the NASA Lewis Icing Research Tunnel which demonstrated that altering the surface tension of the spray can have a dramatic effect on the ice shape. These observations led to the development of a new scaling method in which the scale and reference We are the same.

This method was tested in the IRT using cylinders. Because of a transducer calibration error, the scale We was as much as 28% lower than the reference value. Nevertheless, scale ice shapes closely matched the reference shape when cylinder sizes were scaled by as much as a factor of three. These preliminary results suggest that it may be possible to scale adequately even when We is not held exactly constant. Additional tests are needed to verify this scaling method with a wider range of conditions, with different geometries and with greater size ratios.

Ice-accretion prediction models currently include no consideration of droplet splash. This study demonstrated that consideration of droplet-surface interactions are important to ice accretion and need to be considered not just in scaling methods but in analytical models of ice accretion as well. The Reynolds and Weber numbers need to be investigated carefully to more fully understand their role in the ice-accretion process.

References

1. Bilanin, A. J., "Proposed Modifications to the Ice Accretion/Icing Scaling Theory," AIAA Paper AIAA-88-0203, January 1988.
2. Anderson, David N.: "Rime-, Mixed- and Glaze-Ice Evaluations of Three Scaling Laws," AIAA 94-0718, January, 1994.

3. Ruff, G.A.: "Analysis and Verification of the Icing Scaling Equations," AEDC-TR-85-30, Vol 1 (Rev), March, 1986.
4. Messinger, B.L., "Equilibrium Temperature of an Unheated Icing Surface as a Function of Airspeed," J. Aeron. Sci. 20 No. 1, Jan. 1953, pp 29-42.
5. Bilanin, A.J. and Teske, M.E.: "Numerical Studies of the Deposition of Material Released From Fixed and Rotary Wing Aircraft," NASA CR-3779, March, 1984.
6. Soeder, Ronald H. and Andracchio, Charles, R., "NASA Lewis Icing Research Tunnel User Manual," NASA TM 102319, June 1990.
7. Ide, Robert F., "Liquid Water Content and Droplet Size Calibration of the NASA Lewis Icing Research Tunnel," NASA TM 102447, Jan 1990.
8. Pruppacher, Hans R. and Klett, James D., *Microphysics of Clouds and Precipitation*, Reidel, Boston, 1980.
9. Ingebo, Robert D., "Liquid Fuel Spray Processes in High-Pressure Gas Flow," NASA TM86944, July, 1985.
10. Gelder, Thomas F. and Lewis, James P.: "Comparison of Heat Transfer from Airfoil in Natural and Simulated Icing Conditions," NACA TN 2480, September, 1951.
11. Langmuir, Irving and Blodgett, Katharine B.: "A Mathematical Investigation of Water Droplet Trajectories," Army Air Forces Technical Report No. 5418, February, 1946.

Table I. Order-of-Magnitude Estimates of Nondimensional Parameters Controlling the Dynamics of Impact of a Droplet With a Film

	Rain	Icing
Δ/δ	0 to ∞	0 to 10
$\rho_w U^2 \delta / \sigma_w / a$	10^3	10^3 to 10^4
$\rho_w U \delta / \mu_w$	10^3	10^3

Table II. Surface Tension of Spray Mixtures

Mixture (Concentrations are by Volume)	σ , dyne/cm		
	9-2-93	12-13-93	12-8-94
Tap Water (20°C)	**	63.1	59.2
IRT Demineralized Water	48.6	65.1	59.8
400:1 IRT Demineralized Water : Liqui-Nox	28.7	**	**
560:1 ^a IRT Demineralized Water : Liqui-Nox	30.2	**	**
200:1 IRT Demineralized Water : Photo-Flo 200	28.4	29.8	**
400:1 IRT Demineralized Water : Photo-Flo 200	28.3	28.6	**
600:1 IRT Demineralized Water : Photo-Flo 600	**	**	29.6

^aEstimated Dilution

**Not Measured

Table III. Examples of Scaling With Constant We

Case	Mode	Diam., cm	T_{sp} , K	T_{tot} , K	V , m/s	δ , μm	LWC , g/m^3	τ , min	M	$Re \times 10^{-5}$	$We_{\delta} \times 10^{-3}$
A	Ref.	15.2	263.7	265.9	67.0	40.8	1.10	42.9	.206	7.73	2.36
	Scale	7.6	263.3	267.2	87.6	23.9	1.30	14.0	.269	4.97	2.36
B	Ref.	15.2	266.5	268.7	67.2	40.8	1.11	42.8	.205	7.62	2.39
	Scale	7.6	266.1	270.0	88.0	23.9	1.26	14.4	.269	4.90	2.39
C	Ref.	15.2	260.9	264.9	89.5	40.7	.884	40.3	.276	10.28	4.18
	Scale	7.6	260.2	267.1	117.4	23.7	.993	13.7	.363	6.54	4.18
D	Ref.	7.6	263.7	265.9	66.7	61.0	.897	26.6	.205	3.85	3.50
	Scale 1	5.1	263.5	266.6	78.1	44.6	.988	13.8	.240	2.98	3.50
	Scale 2	2.5	263.0	268.2	102.3	26.0	1.12	4.6	.315	1.90	3.50
E	Ref.	7.6	266.5	268.7	67.4	58.0	.887	26.7	.206	3.82	3.41
	Scale 1	5.1	266.3	269.4	78.8	42.4	.963	14.0	.241	2.95	3.41
	Scale 2	2.5	265.8	271.1	103.3	24.7	1.05	4.9	.316	1.89	3.41

Table IV. Scaling Test Conditions

Case	Mode	Diam., cm	T_{sp} , K	T_{tot} , K	V , m/s	δ , μm	LWC , g/m^3	τ , min	M	$Re \times 10^{-5}$	$We_{\delta} \times 10^{-3}$
A	Ref.	15.2	263.7	265.9	67.0	40.8	1.10	42.9	.206	7.73	2.36
	Scale	7.6	263.7	267.2	83.9	20.7	1.24	15.4	.258	4.76	1.88
B	Ref.	15.2	266.5	268.7	67.2	40.8	1.11	42.8	.205	7.62	2.39
	Scale	7.6	266.3	269.8	83.9	20.8	1.21	15.7	.256	4.68	1.89
C	Ref.	15.2	260.9	264.9	89.5	40.7	.884	40.3	.276	10.28	4.18
	Scale	7.6	260.4	266.6	111.8	20.7	.964	14.8	.346	6.27	3.32
D	Ref.	7.6	263.7	265.9	66.7	61.0	.897	26.6	.205	3.85	3.50
	Scale 1	5.1	263.4	266.3	76.7	38.5	.960	14.4	.236	2.93	2.92
	Scale 2	2.5	262.9	267.5	95.6	21.3	1.08	5.1	.294	1.79	2.51
E	Ref.	7.6	266.5	268.7	67.4	58.0	.887	26.7	.206	3.82	3.41
	Scale 1	5.1	266.2	269.1	76.2	46.4	.967	14.4	.233	2.86	3.49
	Scale 2	2.5	265.8	270.3	95.3	21.5	1.05	5.3	.292	1.76	2.52

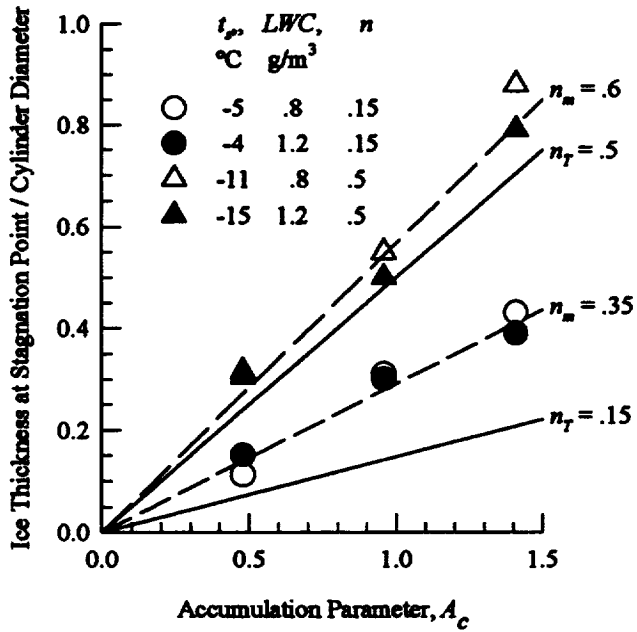


Figure 1. Stagnation Point Ice Accretion on a 2.5-cm-diameter Circular Cylinder. Droplet Velocity, 60 m/s. (Ref. 1)

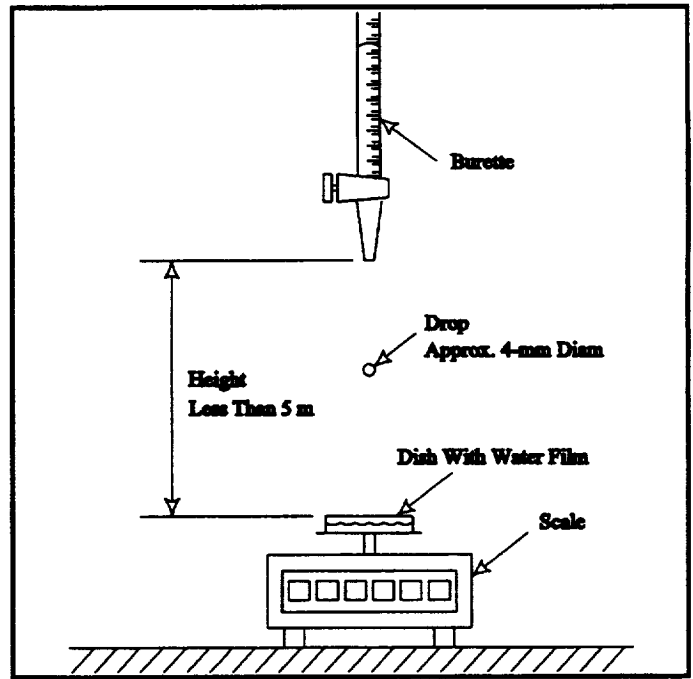


Figure 3. Experimental Setup for Study of Droplet-Film-Impact Dynamics.

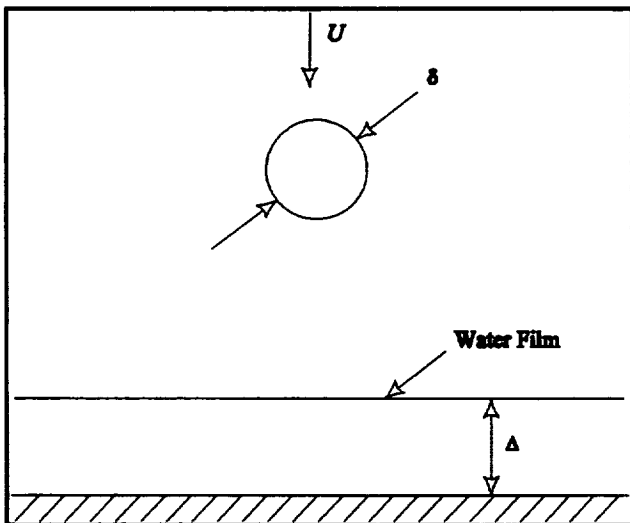


Figure 2. Raindrop Impacting a Shallow Puddle. $\Delta \sim \delta$, U is much less than the acoustic speed in water.

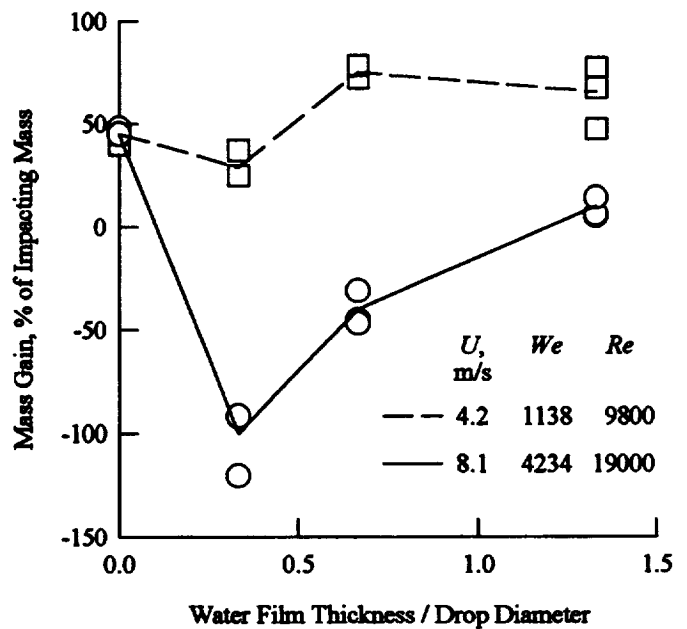


Figure 4. Mass Gain By Water Film Due to Droplet Impact.

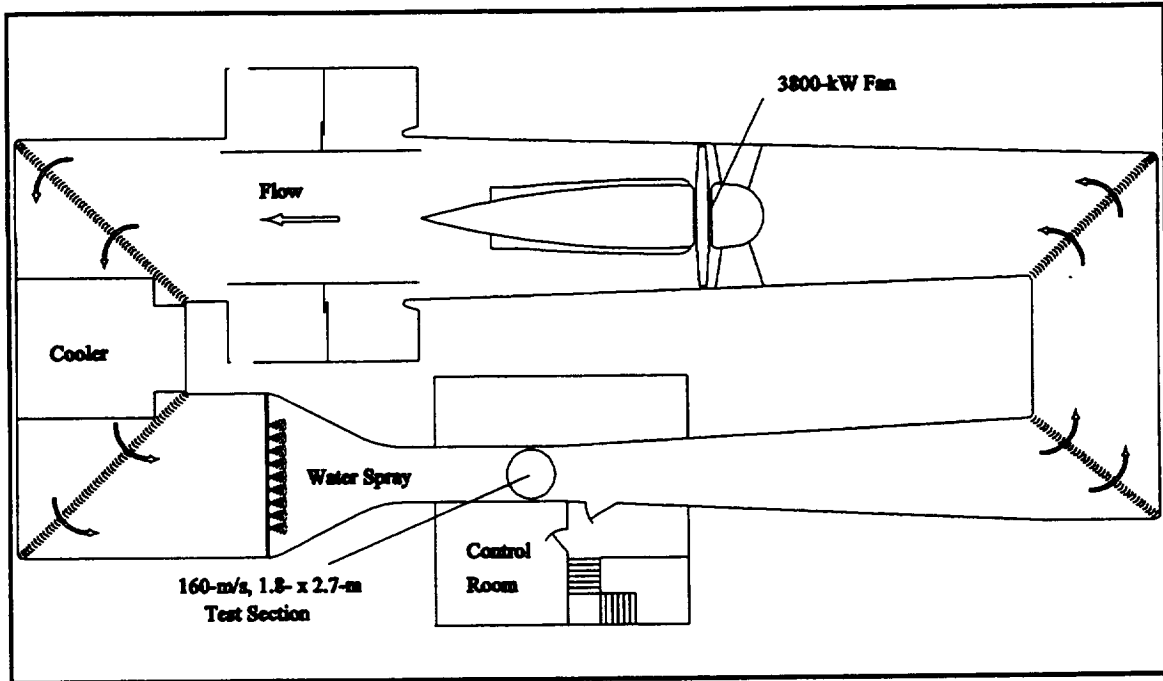


Figure 5. NASA Lewis Icing Research Tunnel (IRT).

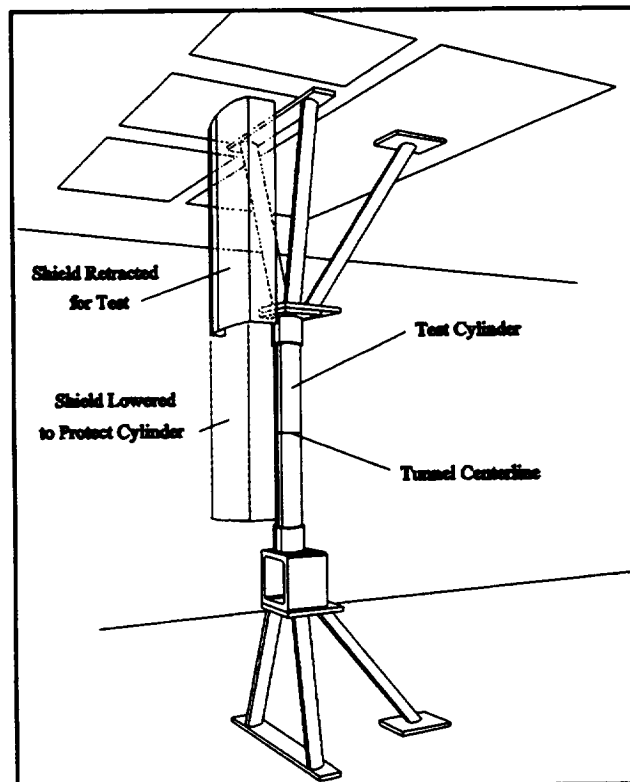


Figure 6. Test Cylinder and Shield Mounted in IRT.

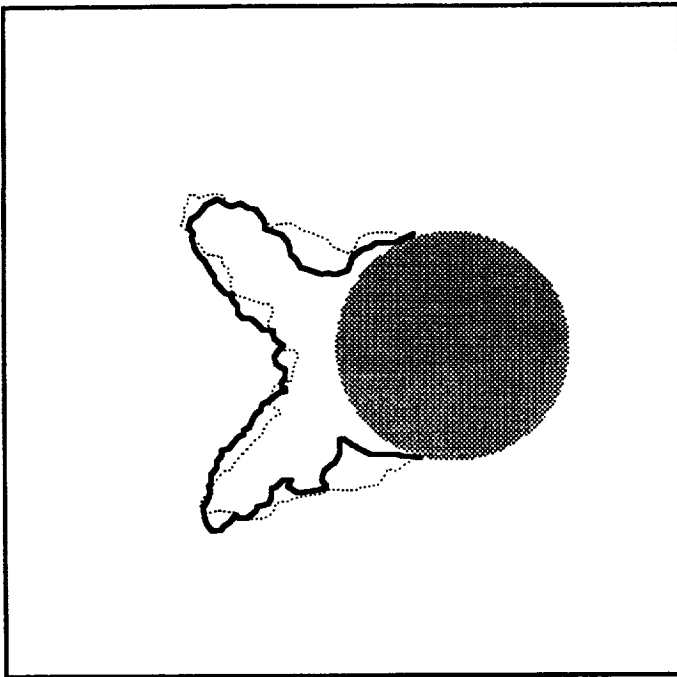
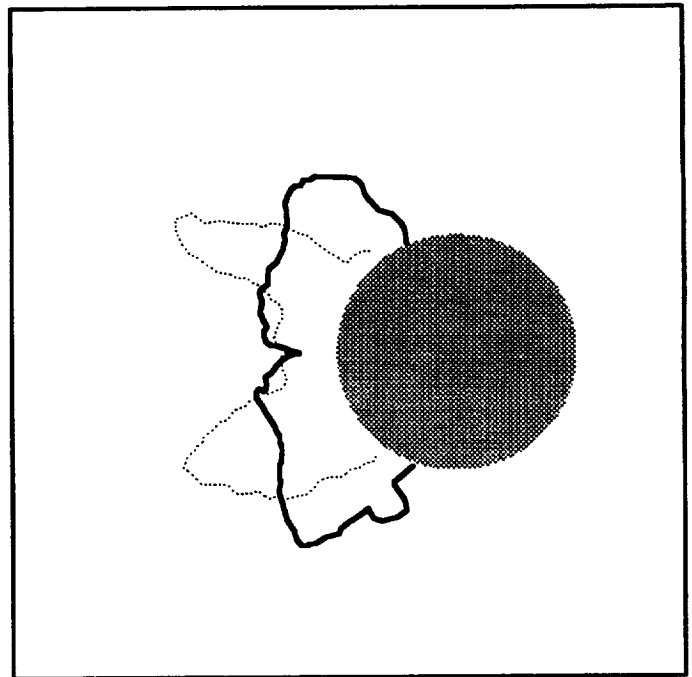
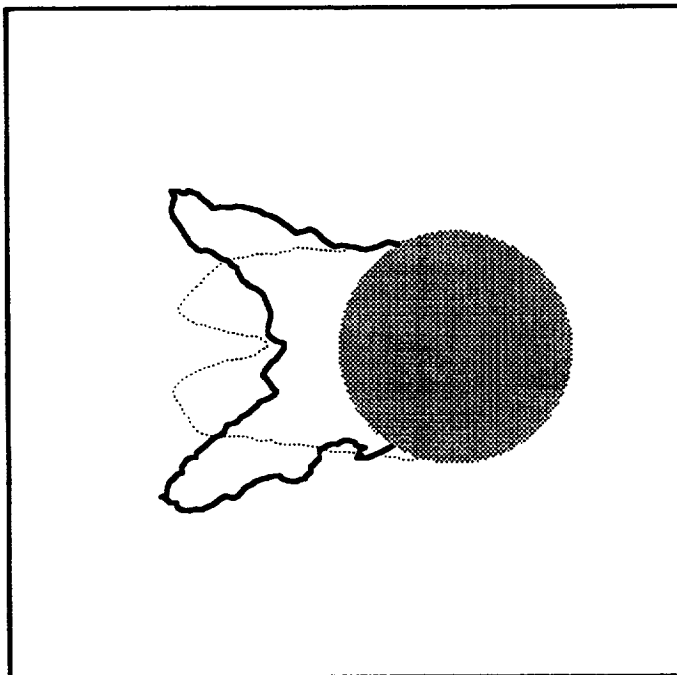


Figure 7. Ice Shape at Model Centerline for Reduced-Surface-Tension Spray. Cylinder diameter, 5 cm; Static Temperature, 266 K; Airspeed, 47 m/s; Median Volume Droplet Diameter, 40 μ m; Liquid-Water Content, 1.17 g/m³; Spray Time, 16 min.
 — 400:1 Water:Liqui-Nox Mixture in Spray Bar System
 600:1 Water:Photo-Flo 600 From Single Nozzle



(b) Static Temperature, 268 K; Airspeed, 47 m/s; Median Volume Droplet Diameter, 40 μ m; Liquid-Water Content, .86 g/m³; Spray Time, 18.7 min.
 — Demineralized Water Spray
 600:1 Water: Photo-Flo 600 From Single Nozzle
Figure 8. (concluded)



(a) Static Temperature, 261 K; Airspeed, 94 m/s; Median Volume Droplet Diameter, 30 μ m; Liquid-Water Content, 1.17 g/m³; Spray Time, 16 min.
 — Demineralized Water Spray
 400:1 Water: Liqui-Nox

Figure 8. Effect of Surfactant on Ice Shape at Model Centerline. Cylinder diameter, 5 cm.

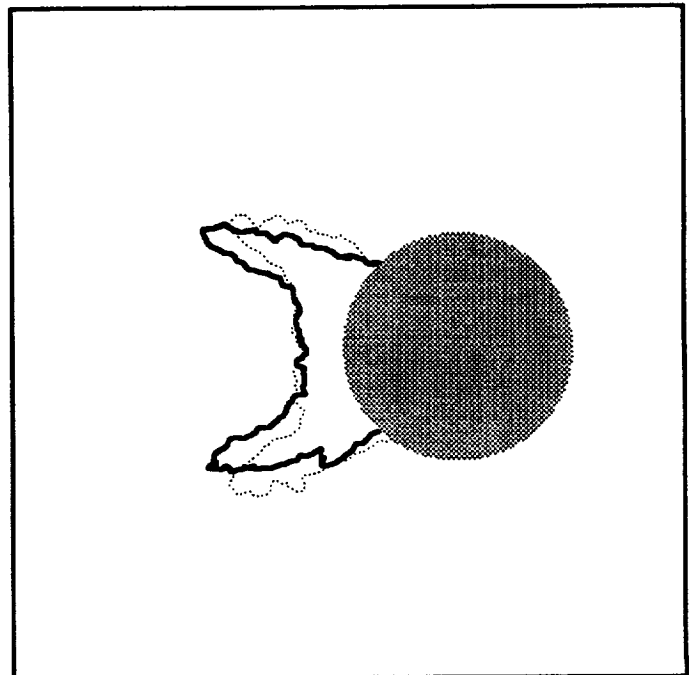


Figure 9. Scaling with Constant Weber Number: Case A, Table IV.

— Reference
 Scale

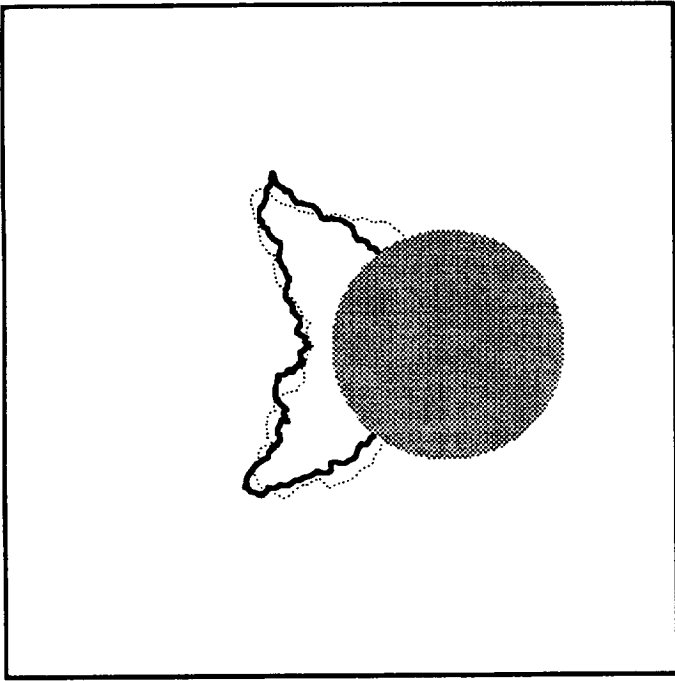


Figure 10. Scaling with Constant Weber Number: Case B in Table IV.

———— Reference
 Scale

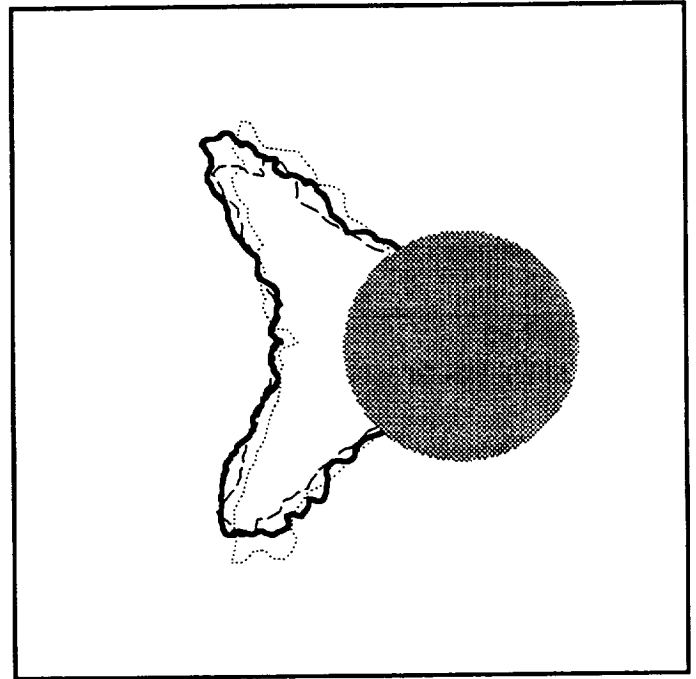


Figure 12. Scaling with Constant Weber Number: Case D in Table IV.

———— Reference
 - - - - Scale 1
 Scale 2

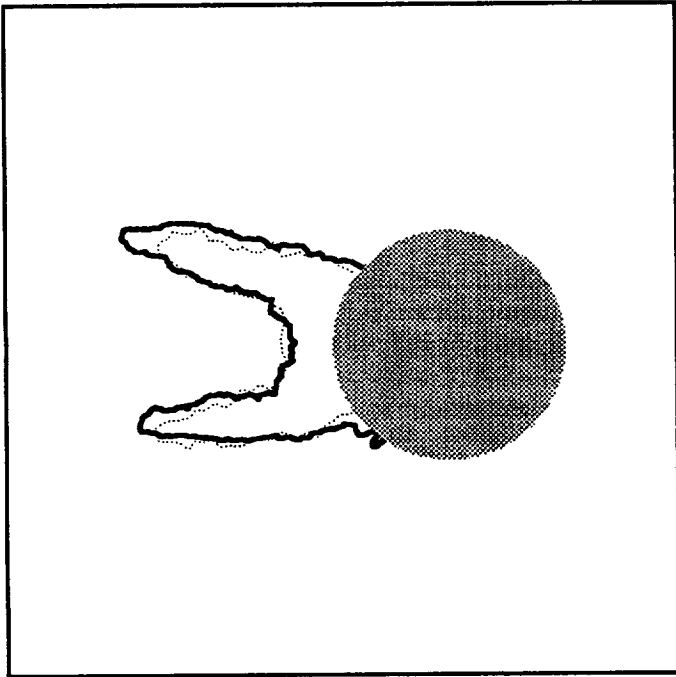


Figure 11. Scaling with Constant Weber Number: Case C in Table IV.

———— Reference
 Scale

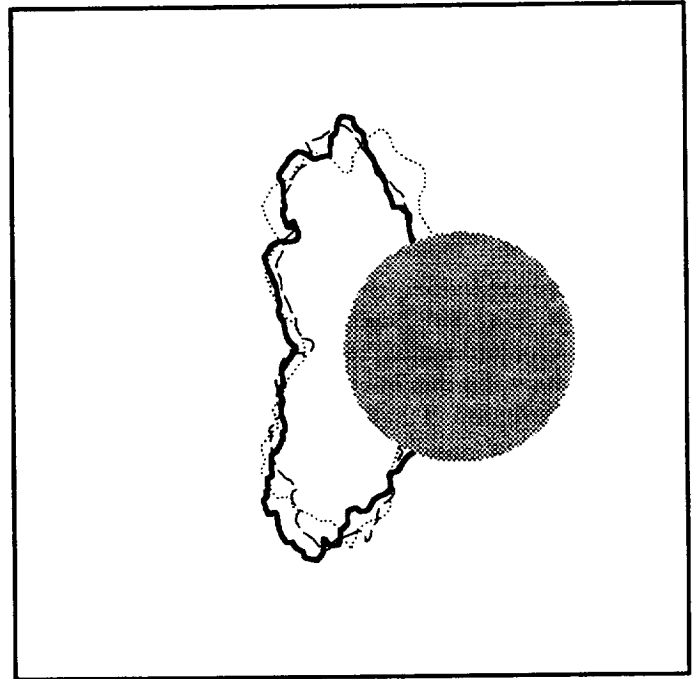


Figure 13. Scaling with Constant Weber Number: Case E in Table IV.

———— Reference
 - - - - Scale 1
 Scale 2

REPORT DOCUMENTATION PAGE

Form Approved
OMB No. 0704-0188

Public reporting burden for this collection of information is estimated to average 1 hour per response, including the time for reviewing instructions, searching existing data sources, gathering and maintaining the data needed, and completing and reviewing the collection of information. Send comments regarding this burden estimate or any other aspect of this collection of information, including suggestions for reducing this burden, to Washington Headquarters Services, Directorate for Information Operations and Reports, 1215 Jefferson Davis Highway, Suite 1204, Arlington, VA 22202-4302, and to the Office of Management and Budget, Paperwork Reduction Project (0704-0188), Washington, DC 20503.

1. AGENCY USE ONLY (Leave blank)	2. REPORT DATE January 1995	3. REPORT TYPE AND DATES COVERED Technical Memorandum	
4. TITLE AND SUBTITLE Ice Accretion with Varying Surface Tension		5. FUNDING NUMBERS WU-505-68-10	
6. AUTHOR(S) Alan J. Bilanin and David N. Anderson		8. PERFORMING ORGANIZATION REPORT NUMBER E-9375	
7. PERFORMING ORGANIZATION NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration Lewis Research Center Cleveland, Ohio 44135-3191		10. SPONSORING/MONITORING AGENCY REPORT NUMBER NASA TM-106826	
9. SPONSORING/MONITORING AGENCY NAME(S) AND ADDRESS(ES) National Aeronautics and Space Administration Washington, D.C. 20546-0001		11. SUPPLEMENTARY NOTES Prepared for the 33rd Aerospace Sciences Meeting and Exhibit sponsored by the American Institute of Aeronautics and Astronautics, Reno, Nevada, January 9-12, 1995. Alan J. Bilanin, Continuum Dynamics, Inc., Princeton, New Jersey, subcontractor to NYMA, Inc. (Subcontract 2705-85, John Deur, Monitor); David N. Anderson, NASA Lewis Research Center. Responsible person, David N. Anderson, organization code 2720, (216) 433-3585.	
12a. DISTRIBUTION/AVAILABILITY STATEMENT Unclassified - Unlimited Subject Category 03 This publication is available from the NASA Center for Aerospace Information, (301) 621-0390.		12b. DISTRIBUTION CODE	
13. ABSTRACT (Maximum 200 words) During an icing encounter of an aircraft in flight, super-cooled water droplets impinging on an airfoil may splash before freezing. This paper reports tests performed to determine if this effect is significant and uses the results to develop an improved scaling method for use in icing test facilities. Simple laboratory tests showed that drops splash on impact at the Reynolds and Weber numbers typical of icing encounters. Further confirmation of droplet splash came from icing tests performed in the NASA Lewis Icing Research Tunnel (IRT) with a surfactant added to the spray water to reduce the surface tension. The resulting ice shapes were significantly different from those formed when no surfactant was added to the water. These results suggested that the droplet Weber number must be kept constant to properly scale icing test conditions. Finally, the paper presents a Weber-number-based scaling method and reports results from scaling tests in the IRT in which model size was reduced up to a factor of 3. Scale and reference ice shapes are shown which confirm the effectiveness of this new scaling method.			
14. SUBJECT TERMS Icing; Scaling; Icing test techniques		15. NUMBER OF PAGES 13	
		16. PRICE CODE A03	
17. SECURITY CLASSIFICATION OF REPORT Unclassified	18. SECURITY CLASSIFICATION OF THIS PAGE Unclassified	19. SECURITY CLASSIFICATION OF ABSTRACT Unclassified	20. LIMITATION OF ABSTRACT