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# TECHNICAL NOTE

## D-397

DATA FROM A STATIC-THRUST INVESTIGATION OF A LARGE-SCALE  
GENERAL RESEARCH VTOL-STOL MODEL IN GROUND EFFECT

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## SUMMARY

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The model was tested at two different elevations with the wing pivot at 1.008 and 2.425 propeller diameters above the ground. The slipstream of the propellers was deflected by tilting the wing and propellers, by deflections of large-chord trailing-edge flaps, and by combinations of flap deflection and wing tilt. Tests were conducted over a range of propeller disk loadings from 7.41 to 29.70 pounds per square foot. Force data for the complete model and pressure distributions for the wing and flaps behind one propeller were recorded and are presented in tabular form without analysis.

## INTRODUCTION

Extensive use of the helicopter has proven the utility of aircraft that are capable of operating without runways. The possible advantages of an airplane which combines both the vertical take-off capabilities of the helicopter and the high cruising speed of conventional airplanes are readily apparent. One possible means of achieving these advantages could be with a tilting wing and propeller or by a combination of flap deflection and wing tilt.

Extensive model investigations (for example, see refs. 1 to 5) have been made of various configurations designed for vertical take-off and landings (VTOL) or for short take-off and landing (STOL). (For a more complete bibliography, see ref. 6.) The model sizes used in the earlier work have prevented obtaining more detailed information on the distribution of aerodynamic loading over the wing and flaps. In addition, the extent to which the model scale might affect the thrust recovery and slipstream turning angles measured was not known. In an effort to provide information of this type, it was decided to test a large-scale general-research VTOL-STOL model.

The present investigation covers the static-thrust characteristics of the model as obtained from tests conducted outdoors at two different

elevations (wing pivot at 1.008 and 2.425 propeller diameters above the ground). The propeller slipstream was deflected by tilting the wing and propeller, by deflecting large-chord trailing-edge flaps, and by using combinations of flap deflection and wing tilt. Performance data were obtained over a range of propeller disk loadings. Pressure distributions were measured over a portion of the wing in order to define the distribution of load on the wing and flaps behind one of the propellers.

#### SYMBOLS

The positive sense of forces, moments, and angles are indicated in figure 1.

b	propeller blade chord, ft	
D	propeller diameter, ft	
h	propeller blade thickness, ft	
R	propeller radius, ft	
r	radius of any propeller blade section, ft	
n	rotational speed, rpm	
c	chord, ft	
F	resultant force, lb	
$F_X$	net longitudinal force (thrust minus drag), lb	
L	lift, lb	
$M_Y$	pitching moment, ft-lb	
T	propeller thrust, total (longitudinal force with wing and flaps undeflected), lb	
z	distance from ground to wing pivot, ft	
$\Delta p$	differential pressure, $p - p_a$	
p	local static pressure	
$p_a$	atmospheric pressure	

$q_s$	slipstream dynamic pressure, $\frac{T}{6\pi R^2}$
$\alpha$	angle of attack, inclination of wing chord above horizontal plane, deg
$\delta_f$	flap deflection, deg
$\theta$	turning angle, inclination of resultant force vector from wing-chord plane
Subscripts:	
55	55-percent-chord flap
30	30-percent-chord flap

#### APPARATUS AND TESTS

A sketch of the model used in these tests is shown in figure 2, and photographs of the model are shown in figures 3, 4, and 5. The airfoil coordinates are given in table I. The geometric characteristics of the model are as follows:

##### Propeller:

Diameter, ft . . . . .	5.0
Solidity (thrust basis) . . . . .	0.1935
Airfoil section . . . . .	NACA 64-OXX

##### Wing:

Span, ft . . . . .	35.0
Chord, ft . . . . .	4.375
Area, sq ft . . . . .	153.125
Airfoil section . . . . .	NACA 63 <sub>2</sub> A215
Pivot, percent c . . . . .	35

##### Flaps:

Span, each wing, ft . . . . .	15.458
Chord, projection of both, percent c . . . . .	55
Chord, projection of rear, percent c . . . . .	30

## Vertical stabilizer:

Span, ft . . . . .	6.0
Chord, ft . . . . .	3.5
Area, sq ft . . . . .	21.0
Airfoil section . . . . .	NACA 0012

## Horizontal stabilizer:

Span, ft . . . . .	16.0
Chord, ft . . . . .	3.0
Area, sq ft . . . . .	48.0
Airfoil section . . . . .	NACA 0012
Pivot, percent c . . . . .	22.86

The model is powered by a single 1,000-horsepower, water-cooled, electric motor located in the fuselage. Power is transmitted to the propellers by means of extension shafts and gear boxes.

The four-bladed propellers have solid aluminum blades. Blade pitch is manually adjustable. Blade form curves are presented in figure 6. The direction of propeller rotation is indicated in figure 2. Rotational speed was measured with signals which were generated by steel vanes on the motor shaft rotating past a magnetic pickup. The output of this pickup was then read on an impulse counter.

The two slotted flaps, 55- and 30-percent wing chord, were mounted on external brackets, as shown in figures 3 and 4. The contours of the flaps are shown in figure 7. The flaps were adjusted manually and were locked in place by pins inserted in the brackets. Flap deflection was measured prior to each run.

The wing was pivoted at the 35-percent-chord station and could be rotated during the test to angles of attack between  $0^\circ$  and  $90^\circ$ . The all-movable horizontal stabilizer was mass-balanced about, and pivoted at, the 22.9-percent-chord station. It was either locked at zero incidence or allowed to float freely, as desired, for each test. Electrical position indicators measured the deflections of the wing and stabilizer.

Four total-pressure tubes (see fig. 8) were installed on the stabilizer chord line and were equally spaced across the right semispan. In order to obtain the average total pressure at the stabilizer, these tubes were manifolded to a single manometer tube. One static-pressure probe was installed at the center of the stabilizer semispan.

The wing and flap behind the center propeller of the right-hand wing panel were fitted with static-pressure orifices. The chordwise and spanwise location of these orifices are given in figure 9. The pressures were indicated on a fluid manometer and photographically recorded.

The surfaces of the wing and flaps had several spanwise joints between wood and metal. It was extremely difficult to maintain a smooth surface over these joints under outdoor conditions where temperature and humidity vary greatly. It is felt, however, that the condition of the surfaces was at least as good as those found on production aircraft. The flaps which were fitted with static-pressure orifices were wrapped in fiber glass in order to maintain an accurate contour.

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The model was mounted on a balance composed of four load cells (figs. 10 and 11). The static weight of the model was supported by automotive-type coil springs in order that more sensitive load cells could be used to measure the aerodynamic loads. The load cells were calibrated in place and thus tares due to the supporting springs and the weight of the model were eliminated. Three vertically oriented load cells, two at the front model supports and one at the rear support, measured lift. A single horizontal load cell at the rear support measured longitudinal force. Pitching moment was calculated from the differences in the restraining forces at the four load cells.

In the low ground position ( $z/D = 1.008$ ), the balance was attached directly to steel plates mounted on a concrete driveway. In the high position ( $z/D = 2.425$ ), the balance and the steel plates were joined by a rigid pipe and channel structure 85.125 inches high, as shown in figures 4 and 5.

The tests were conducted outdoors in an unobstructed area at the Langley Research Center. The nearest structure was a power transformer, which was approximately 25 feet from the left wing tip. The next nearest structure was the Langley helicopter tower, which was approximately 130 feet behind the model. During the early phases of the program, a serious problem of blade pitting developed from recirculation of dirt and sand from the ground areas under the wings and it was necessary to pave a large area beneath the wings of the model.

Most of the data presented were obtained in random-direction winds of from 3 to 6 miles per hour. Approximately 10 percent of the data were obtained at wind speeds below 3 miles per hour.

The electrical power input to the motor was measured throughout the test. Inasmuch as those measurements included large undetermined tares in the power transmission system, the data are not presented.

There were no provisions for direct measurement of the thrust of the propellers. Therefore, all data were reduced by referring the measured forces to a value of propeller thrust defined by the longitudinal force that was measured when both wing and flaps were undeflected and the model was in the high position ( $z/D = 2.425$ ). Therefore the values

of thrust used in data reduction do not reflect the possible effects of flap deflection, angle of attack, ground effect, and the random winds previously discussed. For reference purposes, the values of thrust used in the data reduction, in terms of propeller disk loadings, are given as a function of propeller rotational speed in the following table:

n	Propeller disk loading, lb/sq ft
1,510	7.41
2,085	14.56
2,680	24.81
2,915	29.70

A constant propeller blade pitch angle of  $16.3^\circ$  (at the three-quarter radius) was used throughout the test. The rotational speed of the propellers was held to within  $\pm 20$  revolutions per minute of the desired speed.

The accuracy of the data is believed to be as follows:

Lift, lb . . . . .	$\pm 50$
Longitudinal force, lb . . . . .	$\pm 50$
All angles, deg . . . . .	$\pm 0.2$

The pitching moment is known to contain large errors due to the large moment arms between restraining load cells; consequently, the pitching-moment data given herein should be considered only as a qualitative indication of magnitude.

Inasmuch as the dynamic pressure measured at the floating stabilizer was of the order of 5 percent of the propeller slipstream dynamic pressure, except for the case where the wing and flaps were undeflected, these data are not presented herein.

#### PRESENTATION OF DATA

The data are presented in tabular form without analysis. The force data obtained at  $z/D = 2.425$  are given in table II. The force data obtained at  $z/D = 1.008$  are given in table III. The pressure coefficients measured on the wing and flaps at  $z/D = 2.425$  are given in tables IV to XXII. The pressure coefficients measured on the wing and flaps at  $z/D = 1.008$  are given in tables XXIII to XLIII.



A motion-picture film supplement to this paper has been prepared and is available on loan. A request card form and a description of the film will be found at the back of this paper, on the page immediately preceding the abstract and index pages.

Langley Research Center,  
National Aeronautics and Space Administration,  
Langley Field, Va., March 17, 1960.

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#### REFERENCES

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3. Draper, John W., and Kuhn, Richard E.: Some Effects of Propeller Operation and Location on Ability of a Wing With Plain Flaps To Deflect Propeller Slipstreams Downward for Vertical Take-off. NACA TN 3360, 1955.
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5. Kuhn, Richard E.: Investigation of the Effects of Ground Proximity and Propeller Position on the Effectiveness of a Wing With Large-Chord Slotted Flaps in Redirecting Propeller Slipstreams Downward for Vertical Take-Off. NACA TN 3629, 1956.
6. Kuhn, Richard E.: Semiempirical Procedure for Estimating Lift and Drag Characteristics of Propeller-Wing-Flap Configurations for Vertical- and Short-Take-Off-and-Landing Airplanes. NASA MEMO 1-16-59L, 1959.

TABLE I.- NACA 63<sub>2</sub>A215 AIRFOIL COORDINATES

[Stations and ordinates given in percent of airfoil chord]

Upper surface		Lower surface	
Station	Ordinate	Station	Ordinate
0	0	0	0
.386	1.254	.614	-1.142
.623	1.521	.877	-1.363
1.105	1.959	1.395	-1.717
2.328	2.784	2.672	-2.362
4.804	3.974	5.196	-3.252
7.295	4.863	7.705	-3.891
9.794	5.589	10.206	-4.397
14.804	6.720	15.196	-5.158
19.822	7.547	20.173	-5.687
24.846	8.140	25.151	-6.038
29.873	8.531	30.127	-6.235
34.903	8.719	35.097	-6.271
39.933	8.714	40.067	-6.156
44.963	8.529	45.037	-5.901
49.992	8.188	50.003	-5.528
55.018	7.713	54.982	-5.061
60.041	7.122	59.959	-4.518
65.061	6.428	64.939	-3.918
70.077	5.650	69.923	-3.284
75.090	4.810	74.910	-2.650
80.108	3.924	79.892	-2.054
85.105	2.971	84.895	-1.529
90.074	2.000	89.925	-1.020
95.038	1.016	94.962	-.526
100.00	0	100.00	0

Leading-edge radius: 1.630  
Slope of radius through leading edge: 0.095

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TABLE II.- FORCE MEASUREMENTS AT  $z/D = 2.425$

$\delta_{r,55}$	$\delta_{r,30}$	n	$\alpha$	$\frac{L}{T}$	$\frac{F_X}{T}$	$\frac{F}{T}$	$\theta + \alpha$	Stabilizer locked $\frac{M_y}{TD}$
0	0	1,510	0	0.108	1.004	1.010	6.1	0.026
0	0	2,085	0	.048	.997	.997	2.8	-.073
0	0	2,915	0	.003	1.003	.986	.2	-.068
0	0	1,510	15.0	.297	.956	1.000	17.3	.011
0	0	2,085	15.0	.295	.966	1.009	17.0	.027
0	0	2,915	15.0	.238	.975	1.005	13.7	.012
0	0	1,510	30.0	.616	.850	1.049	36.0	-.186
0	0	2,085	30.0	.600	.875	1.062	34.4	.004
0	0	2,915	30.0	.565	.862	1.031	33.2	-.053
0	0	1,510	45.0	.879	.663	1.101	53.0	.039
0	0	2,085	45.0	.756	.664	1.006	48.7	-.038
0	0	1,510	60.0	.952	.437	1.048	65.3	-.025
0	0	2,085	60.0	.900	.420	.993	65.0	-.092
0	0	2,915	60.0	.897	.487	1.021	61.5	-.040
0	0	1,510	75.0	1.103	.185	1.118	80.5	-.261
0	0	2,085	75.0	1.030	.203	1.049	78.9	-.094
0	0	2,915	75.0	.969	.221	.993	77.2	.025
0	0	1,510	90.0	1.042	-.147	1.052	98.0	-.014
0	0	2,085	90.0	.981	-.089	.985	95.2	.001
0	0	2,915	90.0	.978	-.056	.980	93.3	.040
0	28.5	2,915	0	.254	.958	.991	14.9	-.070
0	28.5	1,510	75	.991	0	.991	90.0	.023
0	28.5	2,085	75	.989	0	.989	90.0	-.059
0	28.5	2,915	75	.920	0	.920	90.0	-.090
0	38.6	2,915	0	.355	.917	.983	21.2	-.134
0	38.6	1,510	68.0	1.036	-.024	1.036	91.3	-.092
0	38.6	2,085	68.0	.998	0	.998	90.0	-.058
0	38.6	2,915	68.0	.948	.015	.948	89.1	-.149
0	49.5	2,915	0	.399	.872	.959	24.6	-.146
0	49.5	1,510	64.5	.907	0	.907	90.0	-.160
0	49.5	2,085	64.5	.951	0	.951	90.0	-.146
0	49.5	2,915	64.5	.930	.015	.931	89.1	-.177
19.8	28.5	2,915	0	.455	.867	.979	27.7	-.218
19.8	28.5	1,510	62.0	.957	-.070	.960	94.2	-.096
19.8	28.5	2,085	62.0	.972	-.060	.974	93.5	-.125
19.8	28.5	2,915	62.0	.940	-.029	.941	91.8	-.136

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TABLE III.- FORCE MEASUREMENTS AT  $z/D = 1.003$  - Continued

$\delta_{f,55}$	$\delta_{f,30}$	n	$\alpha$	$\frac{L}{T}$	$\frac{F_X}{T}$	$\frac{F}{T}$	$\theta + \alpha$	Stabilizer locked $\frac{M_y}{TD}$	Stabilizer free $\frac{M_y}{TD}$
0	38.6	1,510	0	0.230	0.941	0.969	13.8	-0.148	
0	38.6	2,085	0	.223	.951	.977	13.2	-.163	
0	38.6	2,915	0	.247	.927	.960	14.9	-.155	
0	38.6	1,510	0	.293	.938	.983	17.3		-0.239
0	38.6	2,085	0	.263	.936	.973	15.7		-.246
0	38.6	2,915	0	.253	.926	.960	15.3		-.241
0	38.6	1,510	74.0	1.113	.024	1.113	88.8	-.035	
0	38.6	2,085	74.0	1.145	0	1.145	90.0	-.076	
0	38.6	2,915	74.0	1.138	0	1.138	90.0	-.083	
0	38.6	1,510	74.0	1.179	0	1.179	90.0		-.099
0	38.6	2,085	74.0	1.176	0	1.176	90.0		-.058
0	38.6	2,915	74.0	1.130	0	1.130	90.0		-.109
0	49.5	1,510	0	.334	.931	.989	19.7	-.158	
0	49.5	2,085	0	.300	.935	.983	17.8	-.215	
0	49.5	2,915	0	.310	.913	.963	18.8	-.187	
0	49.5	1,510	0	.378	.935	1.008	22.0		-.242
0	49.5	2,085	0	.359	.856	.928	22.7		-.296
0	49.5	2,915	0	.309	.891	.943	19.1		-.262
0	49.5	1,510	74.0	1.106	0	1.106	90.0	-.208	
0	49.5	2,085	74.0	1.050	0	1.050	90.0	-.105	
0	49.5	2,915	74.0	1.094	.006	1.094	89.7	-.097	
0	49.5	1,510	74.0	1.088	0	1.088	90.0		-.102
0	49.5	2,085	74.0	1.109	0	1.109	90.0		-.094
0	49.5	2,915	74.0	1.104	0	1.104	90.0		-.120
19.8	28.5	1,510	0	.500	.849	.985	20.5	-.174	
19.8	28.5	2,085	0	.455	.836	.952	18.6	-.185	
19.8	28.5	2,915	0	.426	.847	.949	16.7	-.195	
19.8	28.5	1,510	0	.469	.845	.967	19.1		-.294
19.8	28.5	2,085	0	.468	.855	.975	18.7		-.255
19.8	28.5	2,915	0	.448	.847	.958	17.9		-.252
19.8	28.5	1,510	69.0	1.082	0	1.082	90.0	-.105	
19.8	28.5	2,085	69.0	1.097	0	1.097	90.0	-.122	
19.8	28.5	2,915	69.0	1.114	0	1.114	90.0	-.129	
19.8	28.5	1,510	69.0	1.086	0	1.086	90.0		-.120
19.8	28.5	2,085	69.0	1.097	0	1.097	90.0		-.141
19.8	28.5	2,915	69.0	1.089	0	1.089	90.0		-.145
19.8	38.6	1,510	0	.531	.833	.988	22.5	-.249	
19.8	38.6	2,085	0	.518	.827	.975	22.1	-.222	
19.8	38.6	2,915	0	.482	.808	.941	20.8	-.245	
19.8	38.6	1,510	0	.544	.827	.989	23.3		-.258
19.8	38.6	2,085	0	.500	.828	.967	21.1		-.259
19.8	38.6	2,915	0	.496	.817	.956	21.3		-.257
19.8	38.6	1,510	64.5	1.082	0	1.082	90.0	-.131	
19.8	38.6	2,085	64.5	1.093	0	1.093	90.0	-.115	
19.8	38.6	2,915	64.5	1.102	0	1.102	90.0	-.148	
19.8	38.6	1,510	64.5	1.119	0	1.119	90.0		-.146
19.8	38.6	2,085	64.5	1.094	0	1.094	90.0		-.186
19.8	38.6	2,915	64.5	1.092	0	1.092	90.0		-.167
19.8	49.5	1,510	0	.531	.826	.981	22.7	-.251	
19.8	49.5	2,085	0	.510	.809	.957	22.2	-.219	
19.8	49.5	2,915	0	.494	.783	.925	22.3	-.244	
19.8	49.5	1,510	0	.580	.818	1.004	25.3		-.278
19.8	49.5	2,085	0	.539	.807	.970	23.7		-.304
19.8	49.5	2,915	0	.518	.786	.941	23.4		-.299

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TABLE III.- FORCE MEASUREMENTS AT  $z/D = 1.008$  - Continued

$\delta_{r,55}$	$\delta_{r,30}$	n	$\alpha$	$\frac{L}{T}$	$\frac{F_X}{T}$	$\frac{F}{T}$	$\theta + \alpha$	Stabilizer locked $\frac{M_y}{TD}$	Stabilizer free $\frac{M_y}{TD}$
19.8	49.5	1,510	62.5	1.073	0	1.073	90.0	-0.138	
19.8	49.5	2,085	62.5	1.115	0	1.115	90.0	-.123	
19.8	49.5	2,915	62.5	1.093	0	1.093	90.0	-.161	
19.8	49.5	1,510	62.5	1.059	0	1.059	90.0		-0.173
19.8	49.5	2,085	62.5	1.051	0	1.051	90.0		-.198
19.8	49.5	2,915	62.5	1.085	0	1.085	90.0		-.187
39.3	28.5	1,510	0	.578	.725	.927	38.6	-.238	
39.3	28.5	2,085	0	.579	.712	.918	39.1	-.238	
39.3	28.5	2,915	0	.545	.697	.885	38.0	-.253	
39.3	28.5	1,510	0	.593	.698	.916	40.3		-.288
39.3	28.5	2,085	0	.573	.711	.913	38.9		-.291
39.3	28.5	2,915	0	.591	.715	.928	39.6		-.281
39.3	28.5	1,510	64.4	1.023	0	1.023	90.0	-.100	
39.3	28.5	2,085	64.4	1.054	-.012	1.054	90.6	-.107	
39.3	28.5	2,915	64.4	1.025	0	1.025	90.0	-.108	
39.3	28.5	1,510	64.4	1.038	0	1.038	90.0		-.150
39.3	28.5	2,085	64.4	1.048	-.012	1.048	90.6		-.145
39.3	28.5	2,915	64.4	1.035	0	1.035	90.0		-.130
39.3	38.6	1,510	0	.605	.684	.913	41.5	-.257	
39.3	38.6	2,085	0	.579	.671	.886	40.8	-.237	
39.3	38.6	2,915	0	.555	.671	.871	39.6	-.176	
39.3	38.6	1,510	0	.650	.633	.908	45.8		-.326
39.3	38.6	2,085	0	.617	.651	.892	43.8		-.309
39.3	38.6	2,915	0	.590	.656	.883	41.9		-.336
39.3	38.6	1,510	61.2	1.003	0	1.003	90.0	-.083	
39.3	38.6	2,085	61.2	.997	0	.997	90.0	-.139	
39.3	38.6	2,915	61.2	1.000	0	1.000	90.0	-.131	
39.3	38.6	1,510	61.2	1.034	0	1.034	90.0		-.129
39.3	38.6	2,085	61.2	1.018	0	1.018	90.0		-.135
39.3	38.6	2,915	61.2	.998	0	.998	90.0		-.162
39.3	49.5	1,510	0	.620	.646	.895	43.8	-.256	
39.3	49.5	2,085	0	.592	.641	.872	42.7	-.260	
39.3	49.5	2,915	0	.563	.632	.847	41.7	-.293	
39.3	49.5	1,510	0	.635	.649	.908	44.4		-.288
39.3	49.5	2,085	0	.601	.635	.874	43.4		-.309
39.3	49.5	2,915	0	.599	.682	.908	41.3		-.277
39.3	49.5	1,510	59.0	1.018	0	1.018	90.0	-.089	
39.3	49.5	2,085	59.0	.990	0	.990	90.0	-.143	
39.3	49.5	2,915	59.0	.987	0	.987	90.0	-.161	
39.3	49.5	1,510	59.0	1.022	0	1.022	90.0		-.148
39.3	49.5	2,085	59.0	.989	0	.989	90.0		-.172
39.3	49.5	2,915	59.0	.960	0	.960	90.0		-.186
59.4	28.5	1,510	0	.532	.609	.809	41.1	-.226	
59.4	28.5	2,085	0	.515	.611	.799	40.1	-.246	
59.4	28.5	2,915	0	.537	.601	.806	41.8	-.241	
59.4	28.5	1,510	0	.525	.608	.803	40.8		-.267
59.4	28.5	2,085	0	.574	.600	.830	43.7		-.237
59.4	28.5	2,915	0	.539	.590	.799	42.4		-.313
59.4	28.5	1,510	59.0	.968	0	.968	90.0	-.089	
59.4	28.5	2,085	59.0	.973	0	.973	90.0	-.099	
59.4	28.5	2,915	59.0	.940	0	.940	90.0	-.126	
59.4	28.5	1,510	59.0	.969	-.023	.969	91.4		-.162
59.4	28.5	2,085	59.0	.968	-.024	.968	91.4		-.169
59.4	28.5	2,915	59.0	.964	0	.964	90.0		-.141

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TABLE III.- FORCE MEASUREMENTS AT  $z/D = 1.008$  - Concluded

$\delta_{f,55}$	$\delta_{f,30}$	n	$\alpha$	$\frac{L}{T}$	$\frac{F_X}{T}$	$\frac{F}{T}$	$\theta + \alpha$	Stabilizer locked $\frac{M_y}{TD}$	Stabilizer free $\frac{M_y}{TD}$
59.4	38.6	1,510	0	0.562	0.569	0.800	44.7	-0.248	
59.4	38.6	2,085	0	.563	.574	.804	44.5	-.248	
59.4	38.6	2,915	0	.562	.595	.819	45.4	-.269	
59.4	38.6	1,510	0	.587	.608	.844	44.0		-0.229
59.4	38.6	2,085	0	.572	.591	.823	44.0		-.265
59.4	38.6	2,915	0	.588	.588	.830	45.1		-.245
59.4	38.6	1,510	58.2	.941	0	.941	90.0	-.327	
59.4	38.6	2,085	58.2	.949	0	.949	90.0	-.107	
59.4	38.6	2,915	58.2	.921	0	.921	90.0	-.134	
59.4	38.6	1,510	58.2	.955	-.013	.956	91.4		-.085
59.4	38.6	2,085	58.2	.953	-.012	.953	90.7		-.077
59.4	38.6	2,915	58.2	.947	-.012	.947	90.7		-.143
59.4	49.5	1,510	0	.599	.574	.830	46.2	-.206	
59.4	49.5	2,085	0	.557	.556	.787	45.1	-.278	
59.4	49.5	2,915	0	.556	.558	.787	44.9	-.254	
59.4	49.5	1,510	0	.579	.568	.811	45.6		-.300
59.4	49.5	2,085	0	.569	.583	.815	44.3		-.284
59.4	49.5	2,915	0	.563	.564	.797	45.0		-.290
59.4	49.5	1,510	57.8	.886	.012	.886	80.3	-.117	
59.4	49.5	2,085	57.8	.921	.006	.921	80.6	-.107	
59.4	49.5	2,915	57.8	.904	.003	.904	80.8	-.127	
59.4	49.5	1,510	57.8	.919	0	.919	90.0		-.093
59.4	49.5	2,085	57.8	.890	-.023	.890	91.5		-.149
59.4	49.5	2,915	57.8	.897	0	.897	90.0		-.157
69.3	28.5	1,510	0	.520	.580	.781	41.7	-.224	
69.3	28.5	2,085	0	.530	.588	.792	42.0	-.230	
69.3	28.5	2,915	0	.481	.593	.763	39.1	-.232	
69.3	28.5	1,510	0	.550	.590	.807	43.0		-.273
69.3	28.5	2,085	0	.564	.603	.827	43.0		-.280
69.3	28.5	2,915	0	.537	.589	.797	42.4		-.282
69.3	28.5	1,510	58.8	.901	0	.901	90.0	-.113	
69.3	28.5	2,085	58.8	.904	0	.904	90.0	-.118	
69.3	28.5	2,915	58.8	.911	0	.911	90.0	-.100	
69.3	28.5	1,510	58.8	.930	0	.930	90.0		-.132
69.3	28.5	2,085	58.8	.929	0	.929	90.0		-.102
69.3	28.5	2,915	58.8	.930	0	.930	90.0		-.119
69.3	38.6	1,510	0	.496	.548	.739	42.2	-.203	
69.3	38.6	2,085	0	.520	.573	.773	42.2	-.222	
69.3	38.6	2,915	0	.500	.558	.750	41.9	-.215	
69.3	38.6	1,510	0	.493	.556	.743	41.6		-.316
69.3	38.6	2,085	0	.517	.571	.771	42.1		-.281
69.3	38.6	2,915	0	.520	.575	.776	42.1		-.256
69.3	38.6	1,510	58.2	.889	-.023	.889	91.5	-.086	
69.3	38.6	2,085	58.2	.902	0	.902	90.0	-.059	
69.3	38.6	2,915	58.2	.887	0	.887	90.0	-.088	
69.3	38.6	1,510	58.2	.872	-.023	.872	91.5		-.165
69.3	38.6	2,085	58.2	.908	-.012	.908	90.8		-.119
69.3	38.6	2,915	58.2	.901	0	.901	90.0		-.119

TABLE IV  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{\rho g}$  OBSERVED ON WING

$\delta_{f,55} = 00.0$   $\delta_{f,30} = 00.0$   $z/D = 2.425$

Tube number	$\alpha = 0.0$						$\alpha =$						Tube number	
	n = 2915			n = 2915			n =			n =				
	Spanwise station		140.5	Spanwise station		140.5	Spanwise station		140.5	Spanwise station		126.0		126.0
1	92.0	110.0	118.0	126.0	92.0	110.0	118.0	126.0	92.0	110.0	118.0	126.0	140.5	1
2	-.244	.369	.777	.571	.777	.571	.777	.571	.777	.571	.777	.571	.777	2
3	.239	-.066	-.107	-.439	-.107	-.439	-.107	-.439	-.107	-.439	-.107	-.439	-.107	3
4	.300	.174	-.235	-.278	-.235	-.278	-.235	-.278	-.235	-.278	-.235	-.278	-.235	4
5	.056	-.114	-.167	-.154	-.167	-.154	-.167	-.154	-.167	-.154	-.167	-.154	-.167	5
6	-.348	-1.039	-.093	-.081	-.093	-.081	-.093	-.081	-.093	-.081	-.093	-.081	-.093	6
7	-.246	-.246	-.126	.388	-.126	.388	-.126	.388	-.126	.388	-.126	.388	-.126	7
8	-.218	-.302	-.042	.077	-.042	.077	-.042	.077	-.042	.077	-.042	.077	-.042	8
9	-.035	-.043	-.225	-.059	-.225	-.059	-.225	-.059	-.225	-.059	-.225	-.059	-.225	9
10	-.056	-.107	.003	.100	.003	.100	.003	.100	.003	.100	.003	.100	.003	10
11	-.056	-.147	-.139	-.017	-.139	-.017	-.139	-.017	-.139	-.017	-.139	-.017	-.139	11
12	-.029	-.077	-.179	-.095	-.179	-.095	-.179	-.095	-.179	-.095	-.179	-.095	-.179	12
13	-.056	-.147	-.042	.093	-.042	.093	-.042	.093	-.042	.093	-.042	.093	-.042	13
14	-.052	-.169	-.174	-.095	-.174	-.095	-.174	-.095	-.174	-.095	-.174	-.095	-.174	14
15	.049	-.056	-.131	-.110	-.131	-.110	-.131	-.110	-.131	-.110	-.131	-.110	-.131	15
16	.036	-.045	-.061	-.036	-.061	-.036	-.061	-.036	-.061	-.036	-.061	-.036	-.061	16
17	-.102	-.140	-.128	-.151	-.128	-.151	-.137	-.137	-.151	-.137	-.137	-.151	-.137	17
18	-.043	-.033	-.112	-.077	-.112	-.077	-.045	-.045	-.112	-.077	-.045	-.045	-.112	18
19	-.046	-.116	-.049	-.035	-.049	-.035	-.005	-.005	-.049	-.035	-.005	-.005	-.035	19
20	.014	-.038	.082	.135	.082	.135	.135	.135	.082	.135	.135	.135	.135	20
21	.072	.112	-.088	.001	-.088	.001	.158	.158	-.088	.001	.158	.158	.158	21
22	.017	-.068	.081	.160	.081	.160	.160	.160	.081	.160	.160	.160	.160	22
23	.052	.001	-.065	.082	-.065	.082	.066	.066	.082	.066	.066	.066	.066	23
24	.059	.017	.012	.031	.012	.031	.000	.000	.031	.000	.000	.000	.000	24
25	.086	.061	.010	.051	.010	.051	.005	.005	.051	.005	.005	.005	.005	25
26	.036	.022	.001	.051	.001	.051	.059	.059	.051	.059	.059	.059	.059	26
27		.056	.045	.077	.045	.077	.086	.086	.077	.086	.086	.086	.086	27
28														28

TABLE V  
PRESSURE COEFFICIENTS  $\frac{C_p}{\rho g}$  OBSERVED ON WING

$\delta_{r,30} = 00.0$

$\delta_{r,50} = 00.0$

$\delta_{r,55} = 00.0$

Tube number	$\alpha = 2915$			$\alpha = 2915$			$\alpha = 2680$			$\alpha = 15.0$			Tube number			
	Spanwise station			Spanwise station			Spanwise station			Spanwise station						
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0		118.0	126.0	140.5
1																.371
2																-.286
3																
4																
5																-.170
6																-.128
7																.339
8																.053
9																-.195
10																.053
11																-.056
12																-.113
13																.064
14																-.064
15																-.180
16																-.064
17																-.028
18																-.191
19																-.028
20																-.003
21																.113
22																-.001
23																.180
24																-.049
25																-.005
26																-.005
27																.051
28																.096



TABLE VI  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_\infty}$  OBSERVED ON WING

$\delta_{f,55} = 00.0$        $\delta_{f,30} = 00.0$        $z/D = 2.425$

Tube number	$\alpha = 0$			$\alpha = 30.0$			$\alpha = 60.0$			$\alpha = 90.0$			Tube number
	Spanwise station			Spanwise station			Spanwise station			Spanwise station			
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	
1													1
2							.507	.291		.254	.247	.342	2
3							.113			-.141	.110	-.226	3
4													4
5							-.081	-.143		.223	-.056	-.245	5
6							-.462	-.896		-.134	-.377	-.150	6
7							-.263	-.318		.557	-.274	-.150	7
8							-.245	-.254		.204	-.257	.566	8
9							-.084	-.143	.179	.009	-.237	.075	9
10							-.084	-.143	.179	.009	-.237	.075	10
11							-.100	-.202	-.166	-.053	-.118	.040	11
12							.004	-.102	-.229	-.072	-.139	-.065	12
13							-.086	-.200	-.004	.191	-.023	-.137	13
14							-.143	-.213	-.216	.040	.023	.050	14
15							-.022	-.072	-.197	-.186	-.141	-.125	15
16							-.022	-.050	-.075	-.066	-.056	-.203	16
17							-.289	-.159	-.047	-.045	-.017	-.054	17
18							-.120	-.136	-.152	-.143	-.268	-.139	18
19							-.052	-.063	-.045	-.056	-.121	-.061	19
20							.079	.081	-.045	.004	-.052	-.001	20
21							-.029	-.045	.143	.195	.087	.189	21
22							.084	.081	-.070	-.068	-.036	-.065	22
23							-.020	-.077	.141	.193	.090	-.069	23
24							-.011	-.009	-.072	-.066	-.021	-.069	24
25							.031	.018	.013	-.004	-.001	-.015	25
26							.084	.072	.022	.013	.030	.005	26
27							.047	.081	.070	.083	.069	.061	27
28							.034	.070	.047	.063	.050	.096	28

TABLE VII  
PRESSURE COEFFICIENTS  $\frac{C_p}{q_\infty}$  OBSERVED ON WING

$\delta_{r,55} = 00.0$        $\delta_{r,30} = 00.0$        $z/D = 2.425$

Tube number	$\alpha = 0$				$\alpha = 60.0$				$\alpha = 60.0$				Tube number
	Spanwise station				Spanwise station				Spanwise station				
	92.0	110.0	118.0	140.5	92.0	110.0	118.0	140.5	92.0	110.0	118.0	140.5	
1													1
2													2
3													3
4													4
5													5
6													6
7													7
8													8
9													9
10													10
11													11
12													12
13													13
14													14
15													15
16													16
17													17
18													18
19													19
20													20
21													21
22													22
23													23
24													24
25													25
26													26
27													27
28													28



TABLE IX  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_\infty}$  OBSERVED ON WING

$\delta_{f,55} = 00.0$   $\delta_{f,30} = 00.0$   $z/D = 2.425$

Tube number	$\alpha =$			$\alpha = 90.0$			$\alpha = 90.0$			Tube number	
	n = 2915			n = 2915			n = 2680				
	Spanwise station			Spanwise station			Spanwise station				
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	
1					.442	.147	.301		.439	.470	1
2				.506	.442	.350			.297	.358	2
3				-.241	-.342	.293	.113.		-.322	-.356	3
4				-.303	-.364	.110	-.204		-.248	-.229	4
5				-.255	-.230	.001	-.161		-.157	-.132	5
6				-.173	-.139	-.208	-.748		-.297	.460	6
7				-.752	.472	-.195	-.362		.058	.127	7
8				-.387	.134	-.212	-.284		-.149	-.018	8
9				-.312	-.018	.005	.000	.149	.056	.146	9
10				.002	.143	-.068	-.098	-.157	-.051	-.041	10
11				-.088	-.045	-.079	-.180	-.229	-.130	-.064	11
12				-.079	-.139	.011	-.034	.007	.060	.121	12
13				.020	.032	-.176	-.180	-.223	-.117	-.066	13
14				.022	-.132	-.094	-.223	-.231	-.202	-.168	14
15				-.098	-.219	-.066	-.085	-.094	-.073	-.060	15
16				.029	-.091	-.005	-.056	-.062	-.047	-.041	16
17				.004	-.054	.005	-.094	.076	-.119	-.130	17
18				.230	.129	-.229	-.096	-.049	-.041	-.015	18
19				-.093	-.086	.045	-.051	-.013	.001	.024	19
20				-.043	-.032	-.045	-.017	-.013	.182	.125	20
21				.086	.091	.085	.104	.094	.053	-.011	21
22				-.013	.129	-.045	-.037	-.075	-.053	-.011	22
23				-.036	.102	-.075	.102	.094	.182	.170	23
24				.102	.177	.093	.102	.072	-.056	-.001	24
25				-.009	-.084	-.007	-.007	-.001	-.022	-.007	25
26				.020	.022	.041	.022	.028	.009	.007	26
27				.038	-.004	.013	.075	.085	.085	.070	27
28				.084	.079	.083	.039	.085	.085	.070	28
				.043	.109	.073	.130	.085	.085	.070	
				.079	.079	.066	.081	.058	.096	.036	



TABLE XI  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{\rho V^2}$  OBSERVED ON WING

$\delta_{r,30} = 38.6$

$\delta_{r,55} = 00.0$

$z/D = 2.425$

Tube number	n = 2915						n = 2915						n = 2085						n = 68.0					
	Spanwise station						Spanwise station						Spanwise station						Spanwise station					
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5
1	.028	.345	.184	.601	.491	.448	.252	.242	.290	.230	.688	.144	.296	.085	.000	.290	.230	.688	.144	.296	.085	.000	.290	.230
2	.095	.070	.117	-.298	-.412	.124	-.105	-.143	-.427	-.070	.081	-.043	-.194	.000	-.070	-.427	-.070	.081	-.043	-.194	.000	-.070	-.427	-.070
3	.074	.151	.072	-.327	-.360	.030	-.187	.091	-.250	.000	-.011	.202	.194	.000	-.070	-.250	.000	-.011	.202	.194	.000	-.070	-.250	-.070
4	.028	.0767	.072	-.157	.125	-.087	-.572	.118	-.194	-.169	.187	-.440	.444	.000	-.169	-.194	-.169	.187	-.440	.444	.000	-.169	-.194	-.070
5	.436	-.263	.115	.341	.516	-.440	-.206	.062	.444	.403	-.366	-.202	.444	.403	.403	.444	.403	-.366	-.202	.444	.403	.403	.444	.403
6	.208	-.068	.115	.100	.184	-.202	-.206	.062	.143	.250	-.175	-.222	.143	.250	.250	.143	.250	-.175	-.222	.143	.250	.250	.143	.250
7	.110	-.068	.115	-.076	.062	-.110	-.089	.062	-.062	.114	-.113	-.062	-.062	.114	.114	-.062	.114	-.113	-.062	-.062	.114	.114	-.062	.114
8	.167	-.068	.115	.174	.313	-.070	.091	.242	.192	.377	.113	.113	.192	.377	.377	.113	.377	.113	.113	.192	.377	.377	.113	.377
9	.007	-.188	.072	-.042	.026	-.070	-.103	.242	-.043	-.043	.113	-.043	-.043	-.043	-.043	-.043	-.043	.113	-.043	-.043	-.043	-.043	-.043	-.043
10	-.030	-.188	-.262	-.155	-.003	-.089	-.223	.242	-.183	-.043	.113	-.183	-.043	-.043	-.043	-.043	-.043	.113	-.183	-.043	-.043	-.043	-.043	-.043
11	.163	.053	.072	.151	.258	-.084	.028	.095	.179	.292	.085	.028	.095	.292	.292	.085	.292	.085	.028	.095	.292	.292	.085	.292
12	.018	-.197	-.241	-.133	.007	-.084	-.221	.095	-.156	-.036	-.206	-.221	-.156	-.036	-.036	-.206	-.221	-.156	-.036	-.036	-.206	-.221	-.156	-.036
13	-.121	-.267	-.275	-.269	-.199	-.175	-.305	-.297	-.324	-.290	-.113	-.305	-.297	-.324	-.290	-.113	-.305	-.297	-.324	-.290	-.113	-.305	-.297	-.324
14	-.039	-.167	-.189	-.157	.127	-.093	-.208	-.209	-.198	-.171	-.113	-.208	-.209	-.198	-.171	-.113	-.208	-.209	-.198	-.171	-.113	-.208	-.209	-.198
15	-.039	-.167	-.189	-.157	.127	-.093	-.208	-.209	-.198	-.171	-.113	-.208	-.209	-.198	-.171	-.113	-.208	-.209	-.198	-.171	-.113	-.208	-.209	-.198
16	.033	-.127	-.146	-.150	-.180	-.143	-.276	-.280	-.206	-.227	.159	-.276	-.280	-.206	-.227	.159	-.276	-.280	-.206	-.227	.159	-.276	-.280	-.206
17	.013	.131	.150	.112	.085	.021	.062	.089	.099	.084	.007	.062	.089	.099	.084	.007	.062	.089	.099	.084	.007	.062	.089	.099
18	.159	.176	.159	.205	.252	.143	.116	.156	.208	.247	.128	.116	.156	.208	.247	.128	.116	.156	.208	.247	.128	.116	.156	.208
19	.250	.248	.263	.320	.381	.234	.202	.219	.328	.379	.222	.202	.328	.379	.379	.222	.202	.328	.379	.379	.222	.202	.328	.379
20	.294	.273	.286	.351	.415	.284	.230	.290	.360	.410	.288	.230	.290	.360	.410	.288	.230	.290	.360	.410	.288	.230	.290	.360
21	.028	-.028	-.163	-.051	.047	.009	-.061	.009	-.080	.012	-.019	-.061	.009	-.080	.012	-.019	-.061	.009	-.080	.012	-.019	-.061	.009	.012
22	.036	-.127	-.273	-.142	-.133	.021	-.198	-.286	-.175	-.080	-.011	-.198	-.286	-.175	-.080	-.011	-.198	-.286	-.175	-.080	-.011	-.198	-.286	-.175
23	.607	-1.019	-1.045	-.978	-.791	-.706	-1.129	-1.126	-1.078	-.885	-.732	-1.129	-1.126	-1.078	-.885	-.732	-1.129	-1.126	-1.078	-.885	-.732	-1.129	-1.126	
24	-.290	-.326	-1.175	-1.184	-.947	-.334	-.347	-.304	-1.204	-1.066	-.261	-.347	-.304	-1.204	-1.066	-.261	-.347	-.304	-1.204	-1.066	-.261	-.347	-.304	-1.066
25	-.207	-.231	-.214	-.146	-.121	-.253	-.234	-.234	-.164	-.173	-.261	-.234	-.234	-.164	-.173	-.261	-.234	-.234	-.164	-.173	-.261	-.234	-.234	-.164
26	-.163	-.106	.015	.030	.009	-.189	-.118	.028	.024	-.034	-.218	-.118	.028	.024	-.034	-.218	-.118	.028	.024	-.034	-.218	-.118	.028	-.034
27	.567	.425	.586	.776	.810	.536	.442	.631	.784	.801	.526	.442	.631	.784	.801	.526	.442	.631	.784	.801	.526	.442	.631	.784
28	.281	.372	.372	.514	.508	.253	.253	.425	.517	.501	.501	.253	.253	.425	.517	.501	.501	.253	.253	.425	.517	.501	.501	.253



TABLE XIII  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_0}$  OBSERVED ON WING

$\delta_{r,55} = 19.6$   $\delta_{r,30} = 28.5$   $z/D = 2.425$

Tube number	$\alpha = 0.0$						$\alpha = -62.5$						$\alpha = 62.5$						Tube number						
	n = 2915						n = 2915						n = 2085							n = 62.5					
	92.0		118.0		126.0		140.5		92.0		118.0		126.0		140.5		92.0			118.0		126.0		140.5	
1	-.037						.659	.487	.105	.320	.714	.486	.608	.196	.242	.206	.206	.099	.028	.291	.534	.620	.620		
2	.108	.323					-.225	-.485	.130	-.057	-.228	-.496	-.028	-.127	-.534	-.620	-.620								
3	.156	-.075					-.182	-.137			-.185	-.173													
4	-.037	-.210					-.160	-.134			-.179	-.148													
5	-.262	-.690					.336	.495			.312	.495													
6	-.085	-.164					.135	.209			.132	.203													
7	.049	.062					.056	.157			.047	.137													
8	.191	.214					.183	.239			.213	.223													
9	.165	.139					.196	.318			.213	.223													
10	.060	-.034					.186	.318			.186	.310													
11	.145	.085					.039	.171			.041	.161													
12	-.205	-.498					.164	.249			.163	.227													
13	-.539	-.827					-.289	-.205			-.302	-.224													
14	-.132	-.271					-.498	-.467			-.729	-.680													
15	-.166	-.271					-.280	-.164			-.250	-.168													
16	.210	.230					.130	-.144			-.287	-.148													
17	.391	.362					.266	.487			.282	.508													
18	.428	.431					.593	.679			.616	.682													
19	.415	.397					.461	.528			.437	.534													
20	-.082	-.119					.444	.504			.445	.513													
21	-.092	.024					-.040	.082			-.034	.095													
22	-.388	-.474					-.082	.092			-.046	.073													
23	-.714	-.719					-.352	-.484			-.370	-.495													
24	-.053	-.076					-.800	-.806			-.819	-.763													
25	.060	.046					-.037	.046			-.093	.049													
26	.589	.471					.681	.733			.686	.741													
27		.323					.481	.497			.486	.502													
28																									



TABLE XIV  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_\infty}$  OBSERVED ON WING

$\delta_{f,55} = 19.8$        $\delta_{f,30} = 38.6$        $z/D = 2.425$

Tube number	$\alpha = 0.0$						$\alpha = 57.8$						$\alpha = 57.8$					
	Spanwise station			Spanwise station			Spanwise station			Spanwise station			Spanwise station			Spanwise station		
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5
1	-.041	.302		.616	.340	-.005	.331		.679	.556	-.079	.556		.329		.737		.737
2	.077	-.064		-.291	-.490	.140	-.095		-.275	-.482	-.275	-.482		-.091		-.243		-.243
3	.094					.185												
4																		
5	-.041	-.194		-.200	-.140	-.033	-.269		-.228	-.140	-.228	-.140		-.220		-.205		-.205
6	-.280	-.613		.357	.510	-.207	-.597		.394	.521	.394	.521		-.699		-.200		-.200
7	-.043	-.100		.160	.228	-.023	-.076		.187	.236	.187	.236		-.144		.109		.109
8	.124	.121		.088	.196	.125	.144		.093	.205	.093	.205		.098		.037		.037
9	.232	.247		.245	.287	.255	.281		.263	.296	.263	.296		.278		.233		.233
10	.189	.168		.168	.368	.189	.166		.210	.398	.210	.398		.174		.182		.182
11	.071	-.001		.060	.196	.054	-.042		.025	.216	.025	.216		-.035		.017		.017
12	.170	.128		.200	.293	.181	.126		.183	.306	.183	.306		.121		.149		.149
13	-.257	-.456		-.420	-.293	-.339	-.603		-.378	-.234	-.378	-.234		-.499		-.359		-.359
14	-.529	-.783		-.722	-.696	-.575	-.974		-.827	-.710	-.827	-.710		-.851		-.785		-.785
15	-.177	-.319		-.285	-.198	-.216	-.398		-.304	-.304	-.304	-.304		-.339		-.283		-.283
16	-.228	-.238		-.263	-.191	-.187	-.367		-.163	-.162	-.163	-.162		-.311		-.081		-.081
17	.370	.238		.264	.575	.251	.275		.304	.624	.304	.624		.266		.235		.235
18	.403	.391		.372	.698	.435	.363		.689	.761	.435	.689		.357		.405		.405
19	.420	.380		.423	.599	.484	.421		.564	.646	.484	.646		.438		.458		.458
20	.420	.380		.446	.582	.478	.402		.529	.624	.478	.624		.423		.491		.491
21	.039	-.009		.083	.094	.033	-.068		.050	.119	.033	.119		-.098		-.043		-.043
22	.090	-.132		-.170	-.075	.072	-.210		-.125	-.044	.072	-.044		-.157		-.116		-.116
23	-.673	-.936		-.841	-.631	-.781	-1.089		-.919	-.638	-.781	-.638		-.947		-.881		-.881
24	-.563	-.584		-.838	-.754	-.669	-.732		-1.075	-.767	-.669	-.767		-.669		-1.026		-1.026
25	-.073	-.141		-.115	-.107	-.084	-.160		-.115	-.072	-.084	-.072		-.126		-.114		-.114
26	.015	.009		.043	.003	.017	.003		.046	.035	.017	.035		.020		.035		.035
27	.524	.514		.645	.794	.636	.540		.862	.886	.636	.862		.577		.645		.645
28		.350		.452	.596		.390		.628					.430		.595		.595

TABLE XV  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_0}$  OBSERVED ON WING

$\delta_{f,50} = 19.8$

$\delta_{f,50} = 49.5$

$\delta_{f,50} = 19.8$

$\delta_{f,50} = 19.8$

Tube number	$\alpha = 0.0$						$\alpha = -51.9$						$\alpha = 51.9$						Tube number
	Spanwise station						Spanwise station						Spanwise station						
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5		
1	.166	.281	.256	.532	.307	.264	.385	.256	.417	.264	.417	.264	.444	.296	.183	.627	.370		
2	.115	.079	.224	-.366	-.601	-.617	.102	-.098	-.408	-.617	-.408	.102	.062	-.085	.183	-.315	-.573		
3																			
4																			
5	-.105	-.235	-.249	-.243	-.217	-.224	-.134	-.264	-.258	-.224	-.258	-.134	-.136	-.237	-.226	-.222	-.187		
6	-.264	-.484	.409	.609	.581	.581	-.313	-.389	.485	.621	.485	-.313	-.276	-.467	.432	.596	.596		
7	-.018	-.024	.201	.201	.290	.290	-.028	.017	.231	.315	.315	-.028	-.015	-.003	.218	.280	.280		
8	.139	.162	.118	.118	.245	.245	.148	.178	.129	.267	.267	.148	.163	.194	.128	.253	.253		
9	.234	.269	.262	.262	.307	.307	.262	.258	.269	.267	.269	.262	.280	.284	.276	.335	.335		
10	.196	.179	.241	.241	.405	.405	.178	.165	.239	.331	.331	.178	.198	.191	.253	.428	.428		
11	.049	-.015	.041	.041	.201	.201	.018	-.039	.030	.208	.208	.018	.038	-.027	.050	.218	.218		
12	.175	.139	.115	.115	.309	.309	.159	.191	.085	.195	.195	.159	.175	.151	.116	.327	.327		
13	-.296	-.530	-.596	-.420	-.313	-.313	-.431	-.573	-.450	-.351	-.351	-.431	-.362	-.526	-.568	-.300	-.300		
14	-.707	-.896	-.928	-.903	-.858	-.858	-.777	-.934	-.919	-.904	-.904	-.777	-.717	-.904	-.880	-.814	-.814		
15	-.200	-.367	-.409	-.343	-.271	-.271	-.281	-.400	-.416	-.353	-.353	-.281	-.245	-.381	-.397	-.319	-.319		
16	.170	.334	.372	.372	.254	.254	.226	.372	.397	.271	.271	.226	.233	.374	.388	.206	.206		
17	.247	.251	.264	.388	.656	.656	.245	.245	.393	.748	.748	.245	.261	.276	.292	.405	.405		
18	.452	.341	.407	.679	.735	.735	.484	.321	.691	.788	.788	.484	.475	.362	.483	.783	.795		
19	.456	.411	.428	.588	.643	.643	.454	.381	.596	.691	.691	.454	.444	.421	.424	.627	.709		
20	.458	.384	.424	.549	.622	.622	.467	.349	.556	.661	.661	.467	.448	.389	.448	.557	.654		
21	.209	.167	.069	.198	.277	.277	.218	.144	.180	.286	.286	.218	.187	.156	.156	.214	.303		
22	-.088	-.394	-.577	-.405	-.377	-.377	-.108	-.443	-.577	-.402	-.402	-.108	-.097	-.459	-.568	-.397	-.354		
23	-.845	-1.347	-1.466	-1.303	-1.011	-1.011	-.934	-1.500	-1.531	-1.084	-1.084	-.934	-1.028	-1.459	-1.021	-1.213	-.993		
24	-.279	-.294	-1.017	-.915	-.700	-.700	-.302	-.338	-.987	-.763	-.763	-.302	-.421	-.424	-1.184	-1.021	-.686		
25	-.192	-.234	-.247	-.190	-.183	-.183	-.218	-.231	-.176	-.243	-.243	-.218	-.233	-.183	-.167	-.128	-.159		
26	-.130	-.088	-.043	-.017	-.039	-.039	-.051	-.052	-.015	-.032	-.032	-.051	-.121	-.011	-.043	-.015	-.043		
27	.662	.558	.686	.884	.888	.888	.653	.524	.653	.915	.915	.653	.596	.573	.709	.943	.958		
28		.400	.520	.692	.673	.673	.490	.370	.718	.737	.737	.490	.370	.613	.561	.736	.732		

TABLE XVI

PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_s}$  OBSERVED ON WING

$\delta_{f,55} = 39.3$   $\delta_{f,30} = 28.5$   $z/D = 2.425$

Tube number	$\alpha = 0.0$						$\alpha = 9.3$						$\alpha = 49.3$					
	Spanwise station						Spanwise station						Spanwise station					
	92.0	110.0	118.0	126.0	140.5		92.0	110.0	118.0	126.0	140.5		92.0	110.0	118.0	126.0	140.5	
1	.182	.291	.213	.579	.314		.262	.241	.476	.323		.499	.254	.514	.409			
2	.151	.062	.192	-.260	-.574		.088	-.079	-.385	-.587		.078	-.094	-.374	-.575			
3	.159					.066						.035						
4																		
5	-.091	-.213	-.205	-.123	-.143		-.123	-.280	-.234	-.205		-.137	-.297	-.241	-.203			
6	-.101	-.469	.345	.345	.568		-.159	.317	-.255	-.283		-.122	-.295	-.267	-.224			
7	.075	-.001	.204	.204	.307		.073	.086	.486	.621		.086	.099	.501	.621			
8	.235	.211	.178	.304	.304		.252	.225	.227	.340		.269	.241	.297	.346			
9	.336	.330	.286	.343	.343		.353	.330	.295	.359		.356	.335	.297	.374			
10	.369	.348	.333	.384	.384		.382	.342	.508	.652		.376	.341	.526	.682			
11	.111	.071	.060	.208	.336		.101	.025	.506	.284		.111	.012	.035	.389			
12	.069	.002	-.058	.049	.112		.068	-.058	.371	.139		.071	-.086	.048	.160			
13																		
14	-.1073	-.1234	-.0916	-.091	-.903		-.177	-.1057	-.1028	-.976		-.178	-.1081	-.1071	-.969			
15	-.293	-.385	-.398	-.241	-.113		-.342	-.465	-.451	-.173		-.295	-.475	-.450	-.171			
16	-.244	-.298	-.304	-.200			-.288	-.362	-.354			-.280	-.376	-.343				
17	.272	.278	.294	.218	.381		.258	.253	.270	.208		.280	.267	.282	-.203			
18	.567	.439	.505	.852	.929		.588	.450	.883	.984		.588	.440	.577	.929			
19	.577	.569	.595	.661	.727		.601	.592	.611	.793		.621	.598	.613	.824			
20	.552	.494	.522	.649	.698		.552	.463	.707	.773		.557	.450	.488	.766			
21	-.094	-.130	-.185	-.039	.055		-.162	-.205	-.234	.053		-.152	-.229	-.213	.066			
22	.163	.019	-.052	.034	.097		-.094	-.023	-.087	.111		.091	-.098	.015	.127			
23	-.452	-.486	-.507	-.381			-.532	-.591	-.572			-.539	-.600	-.458				
24	-.684	-.672	-.758	-.744	-.640		-.836	-.870	-.845	-.647		-.840	-.896	-.875	-.659			
25	-.039	-.084	-.091	-.086	-.066		-.073	-.108	-.109			-.061	-.122	-.101	-.073			
26	.074	.052	.027	.035	.030		.076	.051	.026	.032		.086	.048	.035	.028			
27	.659	.590	.642	.777	.813		.700	.562	.602	.877		.722	.562	.618	.906			
28		.426	.453	.577	.600		.627	.577	.426	.867		.649	.579	.445	.672			

TABLE XVII  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{\rho V^2}$  OBSERVED ON WING

$\alpha = 0.0$   $\alpha = 43.5$   $\alpha = 43.5$   $\alpha = 43.5$   
 $\delta_{r,55} = 39.3$   $\delta_{r,30} = 38.6$   $z/d = 2.425$

Tube number	$\alpha = 0.0$				$\alpha = 43.5$				$\alpha = 43.5$				Tube number
	Spanwise station				Spanwise station				Spanwise station				
	92.0	110.0	118.0	140.5	92.0	110.0	118.0	140.5	92.0	110.0	118.0	126.0	
1	.399	.389	.309	.299	.299	.299	.299	.299	.647	.162	.289	.289	.271
2	.131	-.564	-.586	-.094	-.459	-.659	-.659	-.659	.078	-.111	-.535	-.535	-.640
3	.098	-.218	-.229	.026	-.123	-.254	-.254	-.254	.002	-.309	-.284	-.284	-.261
4	-.164	-.253	-.257	-.182	-.344	-.286	-.286	-.286	-.210	-.144	-.320	-.320	-.294
5	-.106	.593	.585	.534	-.126	.671	.671	.671	-.144	.139	.566	.566	.672
6	.099	.329	.362	.105	.239	.381	.381	.381	.109	.259	.395	.395	.396
7	.274	.345	.366	.290	.248	.386	.386	.386	.304	.337	.274	.274	.396
8	.274	.394	.386	.352	.332	.397	.397	.397	.360	.337	.294	.294	.403
9	.347	.403	.408	.392	.341	.496	.496	.496	.391	.355	.320	.320	.478
10	.371	.356	.330	.392	.341	.496	.496	.496	.391	.355	.320	.320	.478
11	.089	.020	.119	.093	.013	.133	.133	.133	.078	.017	.005	.005	.373
12	.050	-.126	.119	.051	-.061	.133	.133	.133	.028	-.071	.045	.045	.116
13	-.050	-.172	.119	.051	-.061	.133	.133	.133	.028	-.071	.045	.045	.116
14	-.291	-.688	-.982	-.371	-.192	-.101	-.101	-.101	-.1414	-.109	-.172	-.172	-.125
15	-.323	-.810	-.1248	-.454	-.167	-.101	-.101	-.101	-.1414	-.109	-.172	-.172	-.125
16	-.250	-.354	.007	-.361	-.167	-.101	-.101	-.101	-.1414	-.109	-.172	-.172	-.125
17	.281	-.248	-.280	.274	-.458	-.295	-.295	-.295	-.383	-.441	-.299	-.299	-.320
18	.591	.287	.440	.274	.269	.474	.474	.474	-.276	.266	-.436	-.436	.449
19	.574	.577	.924	.584	.599	.871	.871	.871	.543	.436	.601	.601	1.036
20	.574	.577	.748	.591	.556	.714	.714	.714	.573	.556	.584	.584	.864
21	-.024	.574	.703	.572	.430	.800	.800	.800	.566	.437	.728	.728	.800
22	.033	.034	.052	-.080	-.104	.039	.039	.039	-.071	-.094	-.040	-.040	.035
23	-.930	.019	-.104	-.051	-.219	-.125	-.125	-.125	-.043	-.170	-.177	-.177	-.142
24	-.774	-.941	-.108	-.108	-.123	-.086	-.086	-.086	-.1089	-.135	-.0914	-.0914	-.081
25	-.078	-.807	-.621	-.938	-.123	-.086	-.086	-.086	-.1089	-.135	-.0914	-.0914	-.081
26	-.078	-.109	-.102	-.083	-.126	-.112	-.112	-.112	-.073	-.134	-.132	-.132	-.129
27	.711	-.034	.875	.034	.047	.053	.053	.053	.731	.038	.025	.025	.005
28	.685	.685	.682	.754	.399	.769	.769	.769	.731	.378	.746	.746	.789



TABLE XIX  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_0}$  OBSERVED ON WING

$\delta_{r,30} = 26.5$   $z/D = 2.425$

$\delta_{r,55} = 59.4$

Tube number	n = 2915 $\alpha = 0.0$						n = 2915 $\alpha = 39.0$						n = 2085 $\alpha = 36.1$					
	Spanwise station						Spanwise station						Spanwise station					
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	
1	.419	.277		.600	.384	.854	.062						.611	.217		.545	.313	
2	.105	-.098		-.240	-.515	-.046	-.176						.070	-.103		-.346	-.583	
3	.092					-.034							.053					
4				-.238	-.212	-.209	-.284						-.141	-.267		-.240	-.202	
5	-.149	-.278		-.275	-.255	.009	.164						.070	-.068		-.273	-.255	
6	.019	-.272		.431	.577	.196	.248						.252	.232		.543	.657	
7	.201	.157		.318	.586	.383	.284						.424	.303		.384	.442	
8	.371	.333		.320	.440	.383	.284						.427	.303		.348	.485	
9	.395	.437	.353	.391	.491	.383	.351	.408					.429	.419	.449	.384	.505	
10	.412	.448	.346	.407	.448	.415	.374	.418					.444	.429	.454	.429	.593	
11	.396	.386	.378	.407	.572	.415	.374	.407	.418				.424	.358	.412	.659	.806	
12	-.427	-.572	-.833	-.644	.739	-.438	-.417	-.670	.407				-.460	-.591	-.672	-.692	-.649	
13		-.509	-.928	-.802	-.591	-.438	-.982	-.589	.552				-.460	-.591	-.672	-.692	-.649	
14		-.021	-.484	-.446	-.8217	-.396	-.416	-.079	-.670				-.460	-.591	-.672	-.692	-.649	
15		-.384	-.444	-.418	-.418	-.316	-.416	-.429	-.416	-.072			-.159	-.561	-.258	-.488	-.172	
16		-.356	-.433	-.418	-.418	-.316	-.416	-.429	-.416	-.072			-.318	-.495	-.470	-.386		
17		.257	.289	.192	.297	-.276	-.243	-.347	-.338	-.385			-.323	-.343	-.333	-.265	-.310	
18		.631	.650	.863	.920	.212	.281	.277	.189	.341			.267	.219	.285	.219	.346	
19		.692	.689	.789	.855	.670	.558	.607	.960	.979			.710	.603	.626	.914	.995	
20		.638	.584	.752	.794	.651	.574	.577	.778	.892			.758	.626	.626	.844	.914	
21		-.218	-.246	-.125	.009	.651	.450	.618	.720	.828			.697	.535	.644	.778	.844	
22		.127	.012	-.175	.009	-.192	.171	-.203	-.175	-.058			-.166	-.174	-.141	-.086	.005	
23		-.503	-.491	-.435	.104	-.135	.034	-.025	-.042	.035			-.146	-.147	-.027	.025	.101	
24		-.477	-.516	-.725	-.405	-.444	-.333	-.475	-.400	-.555			-.520	-.497	-.492	-.391		
25		-.092	-.100	-.110	-.095	-.368	-.405	-.507	-.405	-.555			-.619	-.657	-.581	-.740	-.611	
26		-.027	-.019	-.009	.005	-.187	-.191	-.167	-.193	-.174			-.060	-.068	-.065	-.075	-.091	
27		.731	.650	.820	.859	-.131	-.140	-.039	-.141	-.115			.045	.063	.005	.055	-.005	
28			.516	.643	.674	.736	.693	.665	.837	.896			.806	.593	.545	.894	.920	
					.707		.392	.497	.678	.719				.460	.545	.743	.763	

TABLE XX

PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_\infty}$  OBSERVED ON WING

$\delta_{r,55} = 59.4$

$\delta_{r,90} = 76.6$

$z/D = 2.425$

Tube number	$\alpha = 0.0$				$\alpha = 76.1$				$\alpha = 39.0$					
	Spanwise station				Spanwise station				Spanwise station					
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0
1	.473	.243	.686	.446	.627	.233	.672	.251	-.394	.566	.177	.355	.307	
2	.099	.114	-.100	-.443	.082	-.106	-.251	-.251	-.394	.063	-.134	-.386	-.558	
3					.066									
4														
5	-.124	-.272	-.216	-.131	-.151	-.253	-.206	-.206	-.200	-.132	-.332	-.249	-.177	
6	.111	-.185	-.356	-.398	.121	-.106	.501	.709	.709	.157	.053	-.284	-.200	
7	.273	.231	.335	.411	.302	.251	.395	.489	.489	.307	.264	.573	.683	
8	.446	.443	.400	.496	.467	.381	.396	.529	.529	.469	.327	.426	.464	
9	.452	.510	.488	.548	.466	.483	.452	.574	.574	.434	.408	.411	.525	
10	.466	.524	.482	.606	.501	.503	.457	.634	.634	.449	.426	.441	.609	
11	.443	.455	.575	.749	.478	.443	.421	.831	.831	.434	.363	.639	.789	
12	.431	.4594	-.726	-.438	-.469	-.344	-.667	-.565	-.565	-.436	-.642	-.802	-.464	
13	-1.260	-1.821	-1.869	-1.755	-1.465	-1.772	-1.807	-1.564	-1.564	-1.239	-1.884	-1.914	-1.254	
14	-.369	-.525	-.502	-.410	-.440	-.491	-.468	-.323	-.323	-.304	-.589	-.571	-.434	
15	-.289	-.374	-.363	-.213	-.344	-.326	-.335	-.216	-.237	-.284	-.421	-.436	-.269	
16	.325	.365	.334	.313	.283	.346	.359	.231	.429	.399	.309	.256	.368	
17	.680	.619	.629	.642	.744	.785	.620	.895	.987	.713	.670	.601	.888	
18	.709	.646	.642	.799	.788	.810	.809	.818	.934	.756	.685	.693	.929	
19	.680	.667	.652	.742	.735	.629	.629	.758	.861	.695	.538	.558	.805	
20	.011	-.055	-.125	-.027	-.060	-.068	-.098	-.003	.095	-.017	-.096	-.167	.104	
21	.088	-.080	-.191	-.074	-.011	-.046	-.156	-.084	-.033	.017	-.127	-.246	-.028	
22	.743	.770	.803	.713	.869	.708	.747	.566	-.642	.815	.827	.909	.756	
23					-.721	-.703	-.714	-.748	-.642	-.713	-.751	-.761	-.624	
24	-.814	-.559	-.788	-.773	-.044	-.023	-.018	-.057	-.057	-.028	-.071	-.076	-.063	
25	-.034	-.028	-.044	-.054	.044	.074	.049	.059	.052	.071	.061	.017	.040	
26	.048	.080	.058	.066	.044	.074	.049	.059	.052	.071	.061	.017	.040	
27	.772	.748	.735	.852	.877	.701	.696	.893	.964	.840	.604	.635	.967	
28	.657	.657	.621	.758	.805	.598	.555	.741	.870	.840	.500	.515	.850	

TABLE XXI

PRESSURE COEFFICIENTS  $\frac{\Delta P}{\rho g}$  OBSERVED ON WING

$\delta_{f,55} = 59.4$

$\delta_{f,30} = 49.5$

$z/D = 2.425$

Tube number	$\alpha = 0.0$				$\alpha = -35.7$				$\alpha = 35.7$				Tube number		
	Spanwise station				Spanwise station				Spanwise station						
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0		118.0	126.0
1	.382	.229	.410	.275	.174	.590	.213	.296	.203	.174	.789	.106	.375	.002	.124
2	.180	.001	-.323	-.176	-.229	-.294	-.110	-.236	-.229	-.607	.050	-.134	.375	-.002	-.613
3	.197	-.125	-.280	-.148	.027	.438	.415	.538	.731	.027	-.010	.314	.609	-.217	-.197
4	-.084	.001	.541	.655	.415	.426	.415	.426	.319	-.203	.175	.154	.431	-.360	-.246
5	.196	.275	.392	.532	.415	.481	.415	.524	.599	.229	.124	.299	.609	.609	.748
6	.349	.351	.433	.586	.503	.459	.503	.524	.510	.319	.352	.360	.431	.431	.520
7	.507	.455	.443	.653	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
8	.504	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
9	.530	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
10	.475	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
11	.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
12	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
13	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
14	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
15	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
16	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
17	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
18	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
19	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
20	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
21	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
22	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
23	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
24	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
25	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
26	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
27	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555
28	-.407	.475	.466	.792	.523	.437	.460	.446	.446	.219	.510	.360	.400	.400	.555





TABLE XXIII  
PRESSURE COEFFICIENTS  $\frac{C_p}{q_\infty}$  OBSERVED ON WING

$\delta_{f,55} = 00.0$        $\delta_{f,30} = 00.0$        $z/D = 1.005$

Tube number	n = 2915				n = 2915				n = 2915				n = 2915				Tube number						
	Spanwise station				Spanwise station				Spanwise station				Spanwise station										
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5			
1	-.758	-.630	.788	.231	.231																	1	
2	.040	.347	-.263	-.502	-.502																		2
3	.008	-.055	-.297	-.243	-.243																		3
4	-.017	-.124	-.220	-.101	-.101																		4
5	-.036	-.098	-.154	-.054	-.054																		5
6	-.031	-.092	-.206	.451	.451																		6
7	-.483	-.374	-.078	.075	.075																		7
8	-.374	-.222	-.350	-.112	-.112																		8
9	-.179	-.049	.028	-.047	-.047																		9
10	-.063	-.115	-.160	-.072	-.072																		10
11	-.068	-.150	-.219	-.111	-.111																		11
12	-.179	-.086	-.081	-.042	-.042																		12
13	-.070	-.158	-.211	-.099	-.099																		13
14	-.101	-.178	-.210	-.184	-.184																		14
15	-.004	-.060	-.089	-.081	-.081																		15
16	-.014	-.055	-.087	-.021	-.021																		16
17	-.306	-.170	-.317	-.230	-.230																		17
18	-.237	-.154	-.259	-.179	-.179																		18
19	-.131	-.095	-.127	-.095	-.095																		19
20	-.004	.104	.056	.126	.067																		20
21	-.033	-.093	-.109	-.062	-.061																		21
22	-.004	-.101	.054	.133	.100																		22
23	-.030	-.095	-.107	-.063	-.062																		23
24	.035	.015	-.028	-.001	-.014																		24
25	.001	-.024	-.047	-.009	-.013																		25
26	.014	.004	-.009	.031	.014																		26
27	-.073	.005	-.017	-.044	-.034																		27
28			-.005	-.036	.007																		28

TABLE XXIV  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_\infty}$  OBSERVED ON WING

$\delta_{r,30} = 00.0$

$\delta_{r,30} = 00.0$

$\delta_{r,30} = 00.0$

$\delta_{r,30} = 00.0$

$\delta_{r,30} = 00.0$

$\delta_{r,30} = 00.0$

Tube number	$\alpha = 15.0$				$\alpha = 15.0$				$\alpha = 15.0$				Tube number		
	Spanwise station				Spanwise station				Spanwise station						
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0		118.0	126.0
1					.372	-.371			.673	-.673	-.244	-.350		.623	.295
2					-.549	.345			-.597	.318	.111	-.317		-.317	-.572
3					-.307	-.073			-.205	-.157	.017	-.066		-.329	-.326
4					-.136	-.120			-.136	-.091	-.026	-.143		-.200	-.161
5					.300	-.864			.300	-.091	-.058	-.122		-.124	-.079
6					.001	-.379			.001	-.303	-.867	-.860		.305	.527
7					-.262	-.274			-.262	-.277	-.363	-.385		.013	.143
8					.080	-.059		.080	.002	-.046	-.009	-.036		-.233	-.040
9					-.140	-.126		-.140	-.046	-.025	-.063	-.110		.034	.163
10					-.196	-.171		-.196	-.106	-.030	-.071	-.146		-.044	-.023
11					-.036	-.118		-.036	.022	.125	-.025	-.099		-.090	-.030
12					-.188	-.164		-.188	-.099	-.028	-.071	-.161		.047	.117
13					-.074	-.184		-.074	-.193	-.139	-.111	-.184		-.080	-.029
14					-.120	-.057		-.120	-.078	-.040	-.111	-.184		-.180	-.136
15					-.012	-.042		-.012	-.070	-.023	-.005	-.057		-.066	-.033
16					-.316	-.192		-.316	-.053	-.023	-.010	-.041		-.029	-.019
17					-.161	-.170		-.161	-.201	-.184	-.318	-.220		-.200	-.190
18					-.071	-.108		-.071	-.112	-.043	-.166	-.167		-.102	-.066
19					.057	-.053		.057	-.033	-.043	-.074	-.110		-.102	-.066
20					-.019	.053		-.019	.113	.055	-.020	.044		-.034	-.014
21					.061	-.074		.061	.129	.182	-.020	-.066		.156	.108
22					-.012	-.050		-.012	-.074	.002	-.020	-.048		-.048	.004
23					.001	-.022		.001	.161	.161	.059	.042		.173	.156
24					.026	-.042		.026	-.071	-.081	-.011	-.071		-.051	-.013
25					.071	-.047		.071	-.011	-.014	-.009	.011		-.009	-.005
26					.029	-.009		.029	-.008	.004	.026	.011		-.005	-.002
27									.049	.053	.064	.053		.046	.046
28									.035	.040	.021	-.036		.033	.038
									.091	.075		-.044		.058	.072





TABLE XXVII  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_\infty}$  OBSERVED ON WING

$\alpha = 60.0$

$\delta_{f,30} = 00.0$

$\delta_{f,55} = 00.0$

Tube number	n = 2915				n = 2680				Tube number	
	$\alpha = 60.0$				$\alpha = 60.0$					
	Spanwise station				Spanwise station					
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5
1					.640	.256	.002		.649	.578
2					-.224	.332			-.232	-.478
3					-.293	.312	-.090		-.299	-.344
4					-.222	.149	-.171		-.218	-.179
5					-.100	.036	-.110		-.106	-.082
6					.314	-.001	-.627		.318	-.485
7					.172	.204	-.147		.169	.205
8					.094	.110	-.061		.090	.142
9					.305	.296	.267		.311	.343
10				.274	.037	.024	.039	.284	.046	.078
11				-.115	.049	-.073	-.073	-.107	-.058	.050
12				.207	.274	.290	.205	.220	.255	.279
13				-.062	.049	.017	-.073	-.085	-.041	.060
14				-.151	-.161	-.033	-.163	-.143	-.074	.060
15				-.047	-.014	-.074	-.064	.014	-.012	-.171
16				-.043	-.017	-.006	-.004	.014	-.006	-.064
17				.081	.075	.099	.067	.103	.024	.034
18				.207	.253	.210	.269	.285	.229	.169
19				.278	.299	.311	.283	.333	.301	.288
20				.272	.258	.357	.295	.367	.370	.355
21				.284	.373	.413	.306	.426	.443	.317
22				.071	.098	.129	.088	.137	.099	.099
23				.302	.373	.415	.332	.432	.459	.394
24				.093	.103	.127	.096	.142	.099	.112
25				.142	.176	.141	.103	.177	.125	.069
26				.155	.231	.249	.226	.231	.198	.142
27				.229	.290	.296	.291	.282	.269	.245
28				.277	.382	.393	.315	.393	.420	.432
				.358	.402	.459	.387	.419	.470	.447

TABLE XXVIII  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{\rho g}$  OBSERVED ON WING

$\delta_{f,55} = 00.0$   $\delta_{f,30} = 00.0$   $z/D = 1.008$

Tube number	$\alpha =$				$\alpha = 75.0$				$\alpha = 75.0$				Tube number		
	n = 2915				n = 2915				n = 2660						
	Spanwise station				Spanwise station				Spanwise station						
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5
1						.393	.113		.583	.467	.385	.232		.466	.444
2						.359	.315		-.231	-.546	.352	.281		-.265	-.556
3						.285	-.092		-.299	-.389	.286	-.122		-.308	-.391
4						.109	-.166		-.215	-.209	.108	-.181		-.215	-.207
5						.026	-.082		-.064	-.039	.023	-.051		-.009	-.049
6						.106	-.566		.312	.532	.086	-.487		.354	.537
7						.091	-.129		.191	.240	.081	-.096		.409	.241
8						.138	.076		.144	.165	.135	.090		.162	.165
9						.333	.291	.300	.350	.353	.339	.303	.288	.364	.357
10						.058	.075	.151	.064	.141	.044	.106	.098	.134	.135
11						.017	-.026	.051	.011	.123	.022	.016	-.027	.101	.119
12						.324	.235	.263	.294	.305	.326	.251	.217	.318	.307
13						.023	-.017	.069	.022	.123	.027	.032	-.014	.103	.120
14						-.036	-.110	.011	-.082	-.051	-.037	-.050	-.060	.024	-.062
15						.091	.067	.169	.101	.126	.091	.120	-.060	.181	.129
16						.125	.143	.228	.116	.174	.119	.190	.172	.134	.185
17						.207	.284	.284	.246	.179	.204	.288	.283	.259	.185
18						.289	.318	.321	.321	.297	.292	.325	.313	.333	.305
19						.265	.341	.358	.362	.367	.278	.348	.354	.373	.367
20						.294	.383	.420	.448	.386	.304	.387	.415	.457	.392
21						.123	.181	.263	.203	.221	.118	.387	.218	.254	.230
22						.306	.387	.412	.458	.436	.323	.389	.414	.462	.435
23						.144	.188	.266	.210	.235	.140	.230	.213	.266	.244
24						.150	.209	.258	.200	.206	.143	.235	.224	.234	.220
25						.213	.271	.260	.234	.226	.212	.271	.261	.224	.230
26						.285	.303	.284	.272	.241	.283	.287	.288	.272	.235
27						.272	.374	.387	.409	.418	.270	.372	.396	.429	.431
28						.358	.358	.387	.427	.415	.349	.349	.391	.431	.425

TABLE XXIX  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_\infty}$  OBSERVED ON WING

$\delta_{f,95} = 00.0$   $\delta_{f,50} = 00.0$   $z/D = 1.008$

Tube number	$\alpha =$				$\alpha = 90.0$				$\alpha = 90.0$				Tube number				
	n = 2915				n = 2915				n = 2680					n = 90.0			
	Spanwise station				Spanwise station				Spanwise station					Spanwise station			
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5		
1	.320	.139	.424	.317	.424	.316	.393	.316	.424	.317	.424	.316	.393	.316	.424	1	
2	.324	.286	-.332	-.318	-.332	.393	.316	.393	-.332	-.318	-.332	.393	.316	.393	-.332	2	
3	.289	-.104	-.318	-.318	-.318	.316	.316	.316	-.318	-.318	-.318	.316	.316	.316	-.318	3	
4	.121	-.157	-.188	-.188	-.188	.130	.130	.130	-.188	-.188	-.188	.130	.130	.130	-.188	4	
5	.042	-.016	.060	.374	.508	.044	.044	.044	.060	.508	.044	.044	.044	.044	-.007	5	
6	-.039	-.565	.179	.374	.508	-.003	-.003	-.003	.179	.508	-.003	-.003	-.003	-.003	.491	6	
7	-.016	-.197	.007	.150	.179	.002	.002	.002	.179	.179	.002	.002	.002	.002	.174	7	
8	.038	-.036	.060	.007	.060	.059	.059	.059	.060	.060	.059	.059	.059	.059	.056	8	
9	.244	.224	.245	.263	.245	.271	.271	.271	.245	.245	.271	.271	.271	.271	.257	9	
10	.045	.116	.120	.172	.120	.051	.051	.051	.120	.120	.051	.051	.051	.051	.123	10	
11	.035	.051	.104	.135	.104	.033	.033	.033	.104	.104	.033	.033	.033	.033	.095	11	
12	.244	.188	.236	.236	.236	.268	.268	.268	.236	.236	.268	.268	.268	.268	.231	12	
13	.026	.074	.144	.148	.104	.032	.032	.032	.148	.104	.032	.032	.032	.032	.095	13	
14	-.010	.004	.108	.101	.007	-.016	-.016	-.016	.101	.007	-.016	-.016	-.016	-.016	.033	14	
15	.110	.169	.207	.207	.132	.118	.118	.118	.207	.132	.118	.118	.118	.118	.190	15	
16	.147	.227	.249	.249	.179	.158	.158	.158	.249	.179	.158	.158	.158	.158	.201	16	
17	.139	.197	.192	.192	.192	.187	.187	.187	.192	.192	.187	.187	.187	.187	.071	17	
18	.239	.255	.227	.227	.227	.239	.239	.239	.227	.227	.239	.239	.239	.239	.178	18	
19	.245	.304	.293	.293	.293	.231	.231	.231	.293	.293	.231	.231	.231	.231	.266	19	
20	.274	.342	.342	.342	.338	.277	.277	.277	.338	.338	.277	.277	.277	.277	.351	20	
21	.161	.263	.291	.266	.224	.170	.170	.170	.266	.224	.170	.170	.170	.170	.264	21	
22	.298	.345	.339	.339	.361	.286	.286	.286	.339	.339	.286	.286	.286	.286	.376	22	
23	.174	.266	.291	.291	.235	.180	.180	.180	.291	.235	.180	.180	.180	.180	.277	23	
24	.179	.273	.257	.257	.213	.194	.194	.194	.257	.213	.194	.194	.194	.194	.254	24	
25	.254	.301	.266	.266	.245	.265	.265	.265	.245	.245	.265	.265	.265	.265	.274	25	
26	.318	.333	.304	.304	.271	.322	.322	.322	.304	.271	.322	.322	.322	.322	.293	26	
27	.266	.355	.355	.355	.358	.238	.238	.238	.355	.358	.238	.238	.238	.238	.343	27	
28		.357	.380	.398	.398	.364	.364	.364	.398	.398	.364	.364	.364	.364	.378	28	



TABLE XXX  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_\infty}$  OBSERVED ON WING

$\alpha = 0.0$   $\alpha = 7.5.0$   $\alpha = 14.0.5$   
 $\delta_{f,55} = 00.0$   $\delta_{f,50} = 28.5$   $\delta_{f,30} = 1.008$

Tube number	n = 2915 $\alpha = 0.0$			n = 2915 $\alpha = 7.5.0$			n = 2915 $\alpha = 14.0.5$			n = 2915 $\alpha = 14.0.5$			Tube number		
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0		118.0	126.0
1	-.316	-.688	.427	.886	.749	.352	.068	.714	.541	.541	.714	.541	.541	.714	.541
2	.356	.427	-.054	-.041	-.399	.416	.342	-.173	-.502	-.502	-.173	-.502	-.502	-.173	-.502
3	.355	-.054	.206	-.157	-.284	.315	-.094	-.207	-.190	-.190	-.207	-.190	-.190	-.207	-.190
4	.051	-.119	.051	-.090	-.077	.033	-.088	-.097	-.089	-.089	-.097	-.089	-.089	-.097	-.089
5	-.308	-1.123	.308	.073	.350	.084	-.628	.294	.531	.531	.294	.531	.531	.294	.531
6	-.209	-.453	.209	-.038	.063	.105	-.139	.176	.236	.236	.176	.236	.236	.176	.236
7	-.127	-.226	.127	-.169	-.024	.165	.071	.116	.170	.170	.116	.170	.170	.116	.170
8	.061	.077	.061	.099	.211	.362	.306	.294	.381	.381	.294	.381	.381	.294	.381
10	.005	-.045	.005	-.012	.036	.035	.088	.094	.094	.094	.030	.094	.094	.030	.094
11	-.007	-.142	.007	-.194	.020	.023	-.028	-.045	.062	.062	-.042	.062	.062	-.042	.062
12	.065	.046	.065	.103	.184	.354	.251	.243	.317	.317	.243	.317	.317	.243	.317
13	-.012	-.141	.012	-.057	.024	.028	-.014	-.029	.069	.069	-.024	.069	.069	-.024	.069
14	-.038	-.216	.038	-.167	-.131	-.023	-.113	-.090	.184	.184	-.090	.184	.184	-.090	.184
15	.000	-.115	.000	-.063	-.062	.110	.076	.093	.019	.019	.093	.019	.019	.093	.019
16	-.059	-.145	.059	-.158	-.076	.152	.174	.184	.022	.022	.184	.022	.022	.184	.022
17	.008	.076	.008	.002	.003	.246	.304	.304	.246	.246	.304	.246	.246	.304	.246
18	.100	.127	.100	.113	.147	.324	.363	.338	.319	.319	.338	.319	.319	.338	.319
19	.183	.185	.183	.196	.261	.300	.326	.386	.450	.450	.386	.450	.450	.386	.450
20	.200	.226	.200	.268	.356	.292	.413	.424	.494	.494	.424	.494	.494	.424	.494
21	-.011	-.137	-.011	-.002	.103	.236	.267	.287	.233	.233	.287	.233	.233	.287	.233
22	.076	-.051	.076	.062	.103	.249	.311	.321	.248	.248	.321	.248	.248	.321	.248
23	-.227	-.360	.227	-.243	-.152	.154	.223	.220	.078	.078	.220	.078	.078	.220	.078
24	-.585	-.711	.585	-.791	-.683	.202	.282	.295	.106	.106	.295	.106	.106	.295	.106
25	-.065	-.088	.065	-.086	-.061	.234	.256	.272	.256	.256	.272	.256	.256	.272	.256
26	.022	.005	.022	.028	.043	.471	.509	.510	.460	.460	.510	.460	.460	.510	.460
27	.336	.422	.336	.617	.667	.471	.509	.510	.460	.460	.510	.460	.460	.510	.460
28	.259	.259	.259	.371	.389	.488	.488	.468	.581	.581	.468	.581	.581	.468	.581

TABLE XXXI  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q}$  OBSERVED ON WING

$\delta_{f,55} = 00.0$   $\delta_{f,30} = 38.6$   $z/D = 1.008$

Tube number	n = 2915						n = 2915						n = 2085						n = 74.0					
	Spanwise station						Spanwise station						Spanwise station						Spanwise station					
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5
1	.242	-.587	.180	.865	.756	.329	.027	.671	.539	.351	.181	.734	.596											
2	.357	-.409	-.066	-.247	-.396	.379	.333	-.196	-.501	.368	.315	-.171	-.502											
3	.200	-.078	-.247	-.193	-.285	.317	-.102	-.287	-.354	.247	-.106	-.282	-.388											
4	.050	-.159	-.193	-.123	-.144	.128	-.181	-.217	-.190	.065	-.179	-.224	-.237											
5	-.230	-1.138	-.123	-.123	-.080	.038	-.093	-.103	-.091	.012	-.080	-.095	-.111											
6	-.162	-.404	.111	.111	.345	.050	-.039	.316	.515	.080	-.611	.277	.512											
7	-.162	-.404	-.010	-.010	.070	.071	-.142	.180	.233	.048	-.149	.176	.227											
8	-.066	-.167	-.115	-.115	.013	.142	.084	.108	.168	.111	.070	.134	.164											
9	.130	.136	.162	.162	.266	.343	.296	.302	.385	.292	.308	.340	.368											
10	.028	-.010	-.026	-.026	.064	.039	.085	.023	.097	.020	.093	.048	.103											
11	.019	-.139	-.099	-.099	.040	.027	-.039	-.072	.067	.012	-.012	-.012	.075											
12	.128	.091	.143	.143	.221	.343	.247	.228	.324	.303	.260	.305	.308											
13	.021	-.127	-.174	-.080	.048	.052	-.029	-.045	-.076	.015	-.005	.000	.075											
14	-.032	-.224	-.219	-.207	-.147	-.017	-.130	-.114	-.158	-.025	-.193	-.123	-.126											
15	-.003	-.107	-.110	-.106	-.068	.119	.059	.055	-.014	.101	.086	.063	-.005											
16	-.040	-.104	.054	.054	.297	.156	.150	.146	.029	.158	.119	.063	.000											
17	.097	.149	.163	.081	.079	.244	.324	.327	.261	.214	.318	.260	.237											
18	.180	.207	.189	.193	.230	.327	.389	.331	.335	.287	.376	.330	.368											
19	.243	.225	.294	.319	.355	.347	.352	.401	.436	.497	.431	.411	.489											
20	.233	.284	.345	.356	.406	.353	.452	.442	.476	.531	.449	.462	.489											
21	.035	.011	.002	.019	.132	.291	.302	.284	.280	.291	.305	.298	.282											
22	.050	-.029	-.016	-.044	-.044	.249	.307	.249	.253	.150	.257	.252	.156											
23	-.496	-.477	-.641	-.755	-.521	.045	.155	.121	.026	-.171	.186	.088	-.131											
24	-.372	-.189	-.346	-.819	-.771	.254	.274	.282	.233	.128	.292	.262	.143											
25	-.128	-.141	-.128	-.078	-.074	.254	.245	.233	.250	.251	.257	.255	.262											
26	-.060	-.083	-.043	-.043	.035	.237	.232	.276	.249	.263	.267	.262	.262											
27	.408	.535	.487	.767	.804	.647	.516	.569	.769	.865	.550	.762	.840											
28	.349	.349	.346	.467	.503	.531	.531	.490	.534	.638	.528	.477	.611											



TABLE XXXIII  
PRESSURE COEFFICIENTS  $\frac{C_p}{q_\infty}$  OBSERVED ON WING

$\delta_{f,50} = 19.8$

$\delta_{f,30} = 28.5$

$\delta_{f,55} = 19.8$

Tube number	$\alpha = 0.0$				$\alpha = 6.9.0$				$\alpha = 69.0$				Tube number	
	Spanwise station				Spanwise station				Spanwise station					
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0		118.0
1	-.179	-.407	.195	.804	.511	-.429	.233	.434	.299	.640	.422	.072	.239	
2	.084	.359	.124	-.169	-.488	.104	.255	-.376	-.596	.076	.177	-.513	-.645	
3	.064	-.067	.044	-.264	-.292	-.043	-.097	-.343	-.363	-.011	-.124	-.391	-.502	
4	.023	-.157	.074	-.185	-.154	-.016	-.156	-.214	-.199	-.062	-.160	-.239	-.244	
5	-.091	-.213	.044	-.167	-.153	.016	-.035	-.123	-.167	.031	-.028	-.081	-.199	
6	-.417	-.791	.044	.241	.466	-.262	-.324	.454	.607	-.249	-.184	.522	.542	
7	-.145	-.209	.017	.090	.161	-.005	.065	.242	.295	.000	.062	.249	.283	
8	.015	.044	.017	.017	.123	.159	.197	.143	.242	.158	.177	.122	.230	
9	.173	.195	.207	.207	.230	.276	.287	.269	.295	.237	.247	.285	.290	
10	.148	.124	.178	.178	.320	.261	.256	.272	.402	.242	.234	.285	.366	
11	.042	-.043	.044	.044	.173	.206	.193	.154	.225	.242	.206	.177	.206	
12	.135	.074	.108	.162	.250	.252	.205	.241	.308	.244	.230	.256	.295	
13	-.298	-.504	-.449	-.291	-.222	.042	-.003	-.058	-.237	.117	.091	-.146	-.261	
14	-.612	-.827	-.773	-.712	-.674	.062	.015	-.444	-.668	.143	.141	-.252	-.693	
15	-.221	-.313	-.309	-.235	-.166	.138	.113	.111	-.017	.158	.172	.072	-.098	
16	-.189	-.269	-.274	-.117	-.142	.126	.100	.091	-.007	.156	.155	.062	-.028	
17	.164	.200	.209	.219	.420	.257	.262	.255	.379	.225	.235	.292	.350	
18	.325	.350	.364	.598	.654	.434	.311	.388	.639	.350	.263	.415	.628	
19	.413	.397	.373	.459	.520	.444	.369	.396	.531	.367	.297	.417	.520	
20	.387	.401	.366	.446	.517	.443	.373	.402	.523	.391	.316	.417	.486	
21	-.107	-.132	-.193	-.017	.091	.188	.172	.143	.149	.235	.254	.141	.189	
22	.070	.019	-.089	.065	.124	.288	.210	.179	.208	.314	.249	.170	.220	
23	-.405	-.479	-.487	-.343	-.254	.105	.100	.044	-.058	.167	.143	.021	.000	
24	-.721	-.745	-.787	-.774	-.620	.237	.228	.150	-.232	.254	.232	.139	.014	
25	-.037	-.070	-.117	-.077	-.044	.255	.247	.220	.188	.266	.266	.235	.199	
26	.043	.057	.036	.043	.035	.242	.237	.213	.205	.249	.254	.225	.215	
27	.574	.481	.552	.696	.689	.577	.469	.792	.752	.470	.407	.604	.774	
28	.330	.372	.372	.487	.482	.383	.383	.429	.575	.470	.347	.670	.585	

TABLE XXXIV  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{\rho g}$  OBSERVED ON WING

$\delta_{f,50} = 19.8$

$\delta_{f,30} = 38.6$

$z/D = 1.008$

Tube number	n = 2915						n = 2085						n = 64.5					
	Spanwise station			Spanwise station			Spanwise station			Spanwise station			Spanwise station			Spanwise station		
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5
1	-.038	-.277	.206	.764	.382	.104	.521	.383	.275	.388	.104	.580	.444	.282	.527	.087		
2	.110	-.328	.143	-.091	-.511	-.407	.091	.204	.221	-.348	-.317	.058	.205	.212	-.377	-.536		
3	.018	-.069	.164	-.300	-.282	-.317	-.019	-.109	.221	-.348	-.317	-.058	.205	.212	-.377	-.536		
4	-.038	-.213	.092	-.213	-.145	-.157	-.082	.176	.025	-.233	-.176	-.096	-.164	.004	-.207	-.154		
5	-.124	-.204	.084	-.218	-.147	-.157	-.065	-.150	.161	-.204	-.157	-.101	-.171	.144	-.193	-.144		
6	-.522	-.665	.020	.261	.478	.628	-.233	-.121	.500	-.204	.628	-.220	-.116	.144	.483	.572		
7	-.138	-.125	.084	.084	.159	.322	.069	.140	.264	.500	.628	.084	.154	.169	.256	.304		
8	.079	.107	.020	.020	.138	.271	.254	.228	.166	.264	.322	.084	.154	.169	.256	.304		
9	.213	.228	.206	.211	.241	.271	.288	.281	.295	.166	.271	.287	.251	.282	.314	.321		
10	.164	.143	.143	.171	.334	.438	.252	.226	.276	.276	.438	.220	.224	.212	.299	.432		
11	.085	-.027	-.041	.014	.179	.246	.150	.094	.101	.101	.246	.106	.043	.004	.116	.246		
12	.145	.098	.092	.145	.262	.335	.229	.188	.238	.335	.335	.207	.169	.144	.258	.335		
13	-.355	-.476	-.463	-.363	-.223	-.246	-.156	.294	-.161	-.335	-.246	-.302	-.411	-.447	-.311	-.220		
14	-.663	-.815	-.765	-.785	-.643	-.663	-.159	-.419	-.659	-.178	-.663	-.396	-.635	-.677	-.681	-.551		
15	-.232	-.325	-.320	-.294	-.191	-.147	.031	-.048	-.182	-.178	-.147	-.028	-.140	-.195	-.174	-.130		
16	-.187	-.305	-.291	-.144	-.170	-.104	.040	-.031	-.132	-.091	-.104	.004	-.072	-.135	-.062	-.082		
17	.192	.219	.209	.225	.473	.659	.254	.261	.309	.413	.659	.265	.290	.321	.425	.609		
18	.371	.308	.336	.637	.667	.739	.344	.302	.471	.697	.739	.328	.333	.481	.730	.744		
19	.386	.356	.410	.508	.567	.658	.378	.348	.478	.597	.658	.379	.367	.485	.628	.684		
20	.388	.354	.406	.478	.560	.648	.414	.353	.467	.573	.648	.381	.369	.466	.584	.633		
21	.034	.022	-.132	-.042	.118	.181	.296	.269	.031	.097	.181	.236	.241	.041	.121	.198		
22	.104	-.175	-.247	-.148	-.025	-.039	.351	.078	-.075	-.002	.039	.275	.038	-.074	.007	.048		
23	-.668	-.920	-.901	-.822	-.497	-.497	.426	-.029	-.528	-.479	-.378	.157	-.278	-.510	-.502	-.340		
24	-.393	-.926	-.906	-.905	-.600	-.600	.247	.142	-.415	-.415	-.400	.246	-.116	-.391	-.488	-.308		
25	-.107	-.121	-.104	-.079	-.080	-.080	.272	.250	.144	.153	.045	.268	.222	.161	.113	.048		
26	-.013	-.027	.066	.058	.011	.126	.269	.269	.251	.232	.269	.265	.265	.261	.203	.106		
27	.457	.493	.665	.749	.790	.854	.409	.394	.554	.846	.854	.381	.512	.705	.875	.889		
28			.501	.599	.555	.672			.685	.685	.672		.423	.575	.730	.681		

TABLE XXXV

PRESSURE COEFFICIENTS  $\frac{\Delta P}{\rho g}$  OBSERVED ON WING

$\delta_{r,55} = 19.8$

$\delta_{r,30} = 49.5$

$z/D = 1.008$

Tube number	$\alpha = 0.0$						$\alpha = 62.5$						$\alpha = 62.5$					
	Spanwise station			Spanwise station			Spanwise station			Spanwise station			Spanwise station			Spanwise station		
	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5	92.0	110.0	118.0	126.0	140.5	140.5
1	-.091	-.244	.751	.429	.507	.534	.299	.096	.523	.424	.540	.266	.540	.266	.540	.266	.540	.266
2	.095	.322	-.230	-.521	.082	.167	-.451	-.578	.064	.177	-.402	-.553	.064	.177	-.402	-.553	.064	.177
3	.010	-.070	-.297	-.368	-.015	-.119	-.368	-.299	-.019	-.093	-.331	-.402	-.019	-.093	-.331	-.402	-.019	-.093
4	-.041	-.152	-.223	-.158	-.070	-.179	-.264	-.149	-.064	-.140	-.214	-.150	-.064	-.140	-.214	-.150	-.064	-.140
5	-.118	-.183	-.227	-.155	-.108	-.169	-.218	-.137	-.108	-.165	-.214	-.195	-.108	-.165	-.214	-.195	-.108	-.165
6	-.542	-.611	.247	.446	-.202	-.072	.511	.532	-.172	-.106	.501	.612	-.172	-.106	.501	.612	-.172	-.106
7	-.122	-.078	.099	.158	.109	.175	.275	.335	.125	.170	.291	.331	.125	.170	.291	.331	.125	.170
8	.142	.154	.059	.159	.304	.230	.188	.303	.325	.266	.308	.306	.325	.266	.308	.306	.325	.266
9	.253	.258	.241	.261	.398	.292	.307	.348	.355	.323	.345	.380	.355	.323	.345	.380	.355	.323
10	.182	.174	.200	.191	.254	.218	.228	.283	.256	.246	.242	.242	.256	.246	.242	.242	.256	.246
11	.065	-.007	.174	.030	.112	.067	.018	.283	.123	.071	.022	.160	.123	.071	.022	.160	.123	.071
12	.175	.130	.160	.264	.234	.175	.152	.379	.237	.190	.163	.298	.237	.190	.163	.298	.237	.190
13	-.372	-.446	-.396	-.233	-.344	-.371	-.466	-.227	-.372	-.412	-.311	-.222	-.372	-.412	-.311	-.222	-.372	-.412
14	-.658	-.778	-.811	-.644	-.475	-.574	-.702	-.607	-.518	-.647	-.689	-.620	-.518	-.647	-.689	-.620	-.518	-.647
15	-.904	-.984	-.994	-.894	-.724	-.858	-.922	-.723	-.632	-.842	-.818	-.713	-.632	-.842	-.818	-.713	-.632	-.842
16	-.160	-.280	-.130	-.180	.038	-.030	-.125	-.082	.044	-.056	-.076	-.086	.044	-.056	-.076	-.086	.044	-.056
17	.226	.261	.292	.508	.287	.266	.329	.481	.298	.306	.340	.525	.298	.306	.340	.525	.298	.306
18	.360	.330	.650	.666	.381	.301	.474	.770	.419	.365	.508	.778	.419	.365	.508	.778	.419	.365
19	.411	.386	.566	.601	.422	.347	.519	.644	.449	.412	.538	.693	.449	.412	.538	.693	.449	.412
20	.402	.386	.579	.579	.447	.374	.508	.708	.471	.427	.516	.639	.471	.427	.516	.639	.471	.427
21	.177	.181	.215	.301	.385	.364	.278	.400	.390	.380	.276	.360	.390	.380	.276	.360	.390	.380
22	-.001	-.342	-.340	-.253	.319	.016	-.240	.400	.311	.004	-.266	.402	.311	.004	-.266	.402	.311	.004
23	-.692	-1.142	-1.139	-.692	.199	-.072	-.628	-.423	.190	-.358	-.735	.696	.190	-.358	-.735	.696	.190	-.358
24	-.235	-.805	-.834	-.533	.264	.154	-.242	-.216	.271	.004	-.422	.252	.271	.004	-.422	.252	.271	.004
25	-.146	-.135	-.073	-.111	.278	.154	.170	.123	.286	.227	.140	.103	.286	.227	.140	.103	.286	.227
26	-.085	-.073	.056	-.003	.298	.272	.240	.181	.296	.289	.229	.118	.296	.289	.229	.118	.296	.289
27	.469	.521	.849	.842	.453	.496	.752	.898	.481	.555	.775	.943	.481	.555	.775	.943	.481	.555
28	.469	.469	.676	.622	.453	.397	.628	.746	.496	.434	.644	.792	.496	.434	.644	.792	.496	.434

TABLE XXXVI  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{\rho g}$  OBSERVED ON WING

$\delta_{f,55} = 39.3$        $\delta_{f,30} = 28.5$        $z/D = 1.008$

Tube number	$\alpha = 0.0$						$\alpha = 64.4$						$\alpha = 64.4$						Tube number
	Spanwise station						Spanwise station						Spanwise station						
	92.0	110.0	118.0	126.0	140.5		92.0	110.0	118.0	126.0	140.5		92.0	110.0	118.0	126.0	140.5		
1	.075	-.170	.070	.644	.603	.513	.415	.215	.396	.167	.621	.573	.310	.375					
2	.090	.326	-.184	-.437	-.437	.089	.215	.400	-.400	-.626	.064	.133	-.530	-.585					
3	.087	-.067	-.282	-.308	-.308	.039	-.101	.349	-.375	-.014	.014	-.047	-.353	-.348					
4	.033	-.171	-.216	-.184	-.184	-.011	-.153	-.327	-.209	-.021	-.021	-.102	-.205	-.203					
5	-.127	-.267	-.226	-.208	-.208	.031	-.032	-.140	-.181	.009	.009	-.014	-.137	-.162					
6	-.292	-.372	.211	.331	.331	-.140	.656	.508	.656	-.100	-.100	-.124	.590	.619					
7	-.029	-.040	.126	.144	.144	.112	.151	.315	.370	.114	.114	.153	.349	.358					
8	.192	.212	.146	.198	.198	.289	.245	.245	.367	.296	.296	.282	.305	.394					
9	.327	.338	.281	.286	.286	.367	.347	.314	.377	.358	.358	.360	.356	.364					
10	.346	.352	.323	.377	.444	.408	.368	.498	.498	.382	.382	.387	.640	.671					
11	.093	.052	.030	.157	.260	.293	.263	.325	.656	.523	.282	.277	.406	.430					
12	.069	-.029	-.086	.032	.092	.276	.239	.180	.216	.216	.267	.246	.213	.227					
13	-.838	-1.013	-.984	-.862	-.780	-.051	-.063	-.190	-.572	-.748	-.066	-.090	-.671	-.662					
14	-1.087	-1.297	-1.274	-1.108	-.972	.188	.170	.124	-.317	-.685	.179	.176	.093	-.540					
15	-.314	-.395	-.377	-.314	-.269	.195	.174	.155	.052	-.073	.181	.176	.143	-.026					
16	-.227	-.271	-.275	-.180	-.192	.187	.170	.149	.008	-.002	.186	.176	.275	-.021					
17	.247	.265	.284	.198	.260	.309	.310	.286	.233	.405	.315	.332	.339	.375					
18	.509	.419	.469	.762	.785	.518	.401	.525	.831	.959	.554	.442	.932	.965					
19	.591	.523	.522	.573	.584	.565	.461	.483	.672	.777	.600	.697	.764	.793					
20	.537	.501	.499	.550	.564	.537	.452	.465	.683	.758	.549	.523	.729	.719					
21	-.115	-.097	-.146	-.037	.059	.204	.184	.179	.181	.217	.222	.270	.215	.234					
22	-.129	.046	-.023	.058	.102	.329	.275	.226	.232	.252	.327	.349	.247	.277					
23	-.413	-.405	-.418	-.345	-.258	.094	.111	.082	-.009	-.072	.081	.100	-.078	-.062					
24	-.578	-.506	-.563	-.620	-.503	.119	.116	.091	-.024	-.105	.119	.117	-.086	-.076					
25	.029	.018	-.011	-.035	-.031	.184	.163	.147	.116	.139	.196	.181	.150	.097					
26	.127	.123	.113	.090	.053	.183	.170	.153	.145	.196	.191	.181	.174	.143					
27	.659	.584	.599	.664	.664	.672	.532	.549	.849	.858	.712	.578	.879	.879					
28			.474	.515	.494	.681	.395	.612	.611	.681	.549	.463	.644	.678					

TABLE XXXVII  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_0}$  OBSERVED ON WING

$\delta_{r,55} = 39.3$        $\delta_{r,30} = 38.6$        $z/D = 1.008$

Tube number	$\alpha = 0.0$						$\alpha = 61.2$						$\alpha = 61.2$							
	n = 2915						n = 2915						n = 2085							
	Spanwise station		Spanwise station		Spanwise station		Spanwise station		Spanwise station		Spanwise station		Spanwise station		Spanwise station		Spanwise station			
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5
1	.038	-.021	.643	.489	.509	.450	.176	.430	.176	.728	.635	.222	.255							
2	.091	.280	-.201	-.432	.077	.220	-.366	-.592	.062	.153	.222	-.499	-.602							
3	.041	-.072	-.276	-.271	.016	-.105	-.334	-.334	-.023	-.128	-.387	-.387	-.548							
4	-.014	-.166	-.201	-.159	-.025	-.165	-.220	-.168	-.078	-.176	-.176	-.248	-.212							
5	-.142	-.244	-.208	-.184	-.004	-.062	-.163	-.165	-.007	-.059	-.198	-.198	-.200							
6	-.332	-.413	.233	.350	-.098	-.079	.508	.640	-.086	.004	.554	.649								
7	.015	.058	.138	.172	.161	.189	.334	.393	.148	.183	.334	.377								
8	.268	.260	.192	.238	.354	.255	.271	.396	.348	.255	.265	.364								
9	.362	.366	.310	.313	.406	.362	.327	.402	.363	.329	.329	.377								
10	.371	.380	.416	.483	.441	.388	.530	.711	.389	.358	.336	.559								
11	.111	.078	.192	.279	.299	.258	.311	.418	.286	.231	.186	.329								
12	.083	-.003	.043	.117	.304	.222	.161	.216	.286	.181	.052	.141								
13	.844	-.955	-.814	-.716	-.149	-.149	-.693	-.794	-.081	-.179	-.365	-.769								
14	-1.090	-1.232	-1.047	-.885	-.188	-.123	-.862	-.888	.178	.121	-.042	-.697								
15	-.303	-.380	-.306	-.249	.198	.135	-.004	.098	.186	.193	-.062	-.136								
16	-.211	-.269	-.176	-.184	.154	.127	-.025	.091	.176	.143	-.066	-.047								
17	.288	.295	.255	.319	.365	.325	.265	.288	.346	.301	.305	.425								
18	.497	.438	.759	.763	.487	.392	.850	.926	.480	.370	.898	.932								
19	.543	.524	.603	.619	.552	.451	.717	.807	.535	.422	.745	.805								
20	.511	.513	.503	.560	.625	.458	.694	.792	.597	.423	.712	.740								
21	.024	.018	.014	.085	.625	.458	.694	.792	.597	.423	.712	.740								
22	.076	-.056	-.079	-.052	.421	.238	.084	.121	.365	.367	.181	.234								
23	-.592	-.651	-.610	-.456	.032	.037	-.131	-.259	.415	.238	.073	.090								
24	-.344	-.316	-.643	-.505	.032	.037	-.131	-.259	.055	.040	-.196	-.258								
25	.013	.010	-.030	-.048	.187	.135	.074	.081	.176	.143	-.186	-.284								
26	.078	.096	.117	.084	.166	.149	.101	.101	.179	.152	.119	.121								
27	.626	.596	.647	.721	.634	.536	.900	.648	.618	.513	.922	.910								
28	.492	.492	.584	.564	.634	.421	.721	.428	.750	.391	.740	.743								



TABLE XXXVIII  
PRESSURE COEFFICIENTS  $\frac{C_p}{q_\infty}$  OBSERVED ON WING

$\delta_{t,55} = 39.3$   $\delta_{t,50} = 49.5$   $z/D = 1.008$

Tube number	$\alpha = 0.0$						$\alpha = 59.0$						$\alpha = 59.0$					
	Spanwise station						Spanwise station						Spanwise station					
	92.0	110.0	118.0	126.0	140.5		92.0	110.0	118.0	126.0	140.5		92.0	110.0	118.0	126.0	140.5	
1	.168	.056	.343	.665	.498	.678	.487	.368	.162	.716	.597	.325	.223	.232	.417	.327	.432	.223
2	.090	.281	.351	.182	.408	.078	.184	.404	.605	.057	.161	.367	.465	.614	.379	.327	.432	.465
3	.027	.066	.073	.268	.258	.000	.118	.354	.343	.019	-.123	.196	-.365	-.356	.417	.327	.432	-.365
4	-.035	.158	.084	.192	.150	-.048	-.174	.236	.185	-.068	-.170	.201	-.244	-.203	.417	.327	.432	-.244
5	-.154	.236	.040	.196	.084	-.010	-.003	.194	.678	-.014	-.066	.201	-.189	-.203	.417	.327	.432	-.189
6	-.263	.316	.008	.241	.362	-.062	-.003	.349	.532	.035	.047	.227	.564	.678	.417	.327	.432	.564
7	.077	.113	.073	.180	.198	.192	.194	.287	.417	.394	.265	.325	.417	.417	.327	.432	.432	.417
8	.311	.282	.343	.225	.276	.378	.251	.287	.426	.394	.265	.325	.417	.417	.327	.432	.432	.417
9	.368	.388	.351	.331	.339	.417	.344	.344	.438	.403	.374	.365	.465	.465	.327	.432	.432	.465
10	.388	.392	.351	.440	.519	.465	.383	.344	.358	.438	.403	.374	.465	.465	.327	.432	.432	.465
11	.380	.392	.351	.215	.304	.303	.234	.201	.424	.280	.265	.196	.322	.422	.327	.432	.432	.322
12	.084	.008	.040	.084	.141	.290	.183	.058	.216	.144	.275	.080	.144	.196	.144	.144	.144	.144
13	-.867	-.950	-.901	-.785	-.669	-.159	-.248	-.434	-.828	-.170	-.170	-.372	-.766	-.856	-.372	-.372	-.372	-.766
14	-1.101	-1.221	-1.172	-1.014	-.818	.154	.087	-.118	-.947	.144	.144	-.019	-.683	-.977	-.019	-.019	-.019	-.683
15	-.303	-.370	-.349	-.285	-.228	.165	.110	.077	-.148	.158	.158	.104	-.038	-.182	.104	.104	.104	-.038
16	-.099	-.188	-.266	-.161	-.168	.165	.108	.075	-.062	.156	.156	.095	-.064	-.078	.095	.095	.095	-.064
17	.311	.321	.324	.280	.349	.375	.317	.321	.337	.370	.370	.322	.370	.503	.322	.322	.322	.370
18	.507	.439	.488	.781	.754	.523	.386	.594	.866	.545	.545	.567	.873	.965	.567	.567	.567	.873
19	.538	.518	.552	.630	.641	.565	.442	.561	.754	.581	.581	.522	.754	.854	.522	.522	.522	.754
20	.518	.500	.503	.580	.603	.641	.464	.464	.718	.631	.631	.505	.711	.788	.505	.505	.505	.711
21	.221	.220	.165	.246	.279	.490	.480	.480	.415	.477	.477	.365	.584	.612	.365	.365	.365	.477
22	-.073	.223	.381	-.249	-.227	.301	.084	-.047	-.105	.244	.244	.004	-.085	-.130	.004	.004	.004	-.085
23	-.596	-.798	-.921	-.785	-.544	.006	-.026	-.078	-.245	-.417	-.417	-.059	-.249	-.465	-.059	-.059	-.059	-.249
24	-.130	-.417	-.589	-.375	-.395	.081	.021	-.028	-.188	-.311	-.311	.002	-.187	-.370	.002	.002	.002	-.187
25	-.025	-.017	.326	-.034	-.055	.161	.107	.084	.047	.066	.066	.123	.066	.047	.123	.123	.123	.066
26	.016	.048	.490	.077	.026	.165	.135	.098	.084	.126	.126	.113	.104	.120	.113	.113	.113	.104
27	.626	.594	.550	.754	.767	.681	.538	.679	.923	.794	.794	.645	.968	.970	.645	.645	.645	.968
28	.512	.512	.511	.641	.620	.681	.431	.560	.794	.812	.812	.441	.785	.835	.441	.441	.441	.785

TABLE XXXIX  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_\infty}$  OBSERVED ON WING

$\delta_{f,25} = 59.4$        $\delta_{f,30} = 28.5$        $z/D = 1.008$

Tube number	$\alpha = 0.0$						$\alpha = 59.0$						$\alpha = 59.0$						Tube number
	Spanwise station			Spanwise station			Spanwise station			Spanwise station			Spanwise station						
	92.0	110.0	126.0	140.5	110.0	126.0	140.5	92.0	110.0	126.0	140.5	92.0	110.0	126.0	140.5				
1	.272	.009	.773	.658	.676	.589	.210	.200	.690	.549	.529	.391	.529	.391	1				
2	.119	.351	-.043	-.363	.090	.170	-.429	-.606	.064	.205	-.324	-.539	-.324	-.539	2				
3	.153	-.077	-.245	-.259	.095	-.089	-.352	-.372	.096	-.099	-.299	-.329	-.299	-.329	3				
4	.076	-.171	-.193	-.148	.031	-.142	-.226	-.200	.029	-.141	-.195	-.173	-.195	-.173	4				
5	-.067	-.199	-.188	-.165	.002	-.053	-.143	-.135	.004	-.061	-.131	-.131	-.131	-.131	5				
6	.003	-.372	.171	.361	.139	.073	.576	.468	.185	-.044	.532	.673	.532	.673	6				
7	.180	.166	.216	.241	.280	.265	.418	.468	.301	.250	.396	.440	.396	.440	7				
8	.352	.398	.314	.356	.430	.320	.385	.500	.450	.331	.373	.509	.373	.509	8				
9	.431	.483	.427	.436	.475	.448	.454	.566	.462	.477	.440	.552	.448	.552	9				
10	.452	.502	.432	.489	.507	.466	.472	.572	.485	.490	.455	.448	.448	.448	10				
11	.447	.456	.407	.461	.543	.455	.454	.688	.505	.467	.462	.675	.462	.675	11				
12	-.278	-.377	-.441	-.372	-.116	-.027	-.336	-.345	.121	-.084	-.279	-.346	.121	-.084	12				
13	-.837	-.883	-1.084	-1.084	-.174	-.031	-.286	-.605	-.141	-.096	-.309	-.700	-.141	-.096	13				
14	-.340	-.314	-.768	-.789	.137	.149	.086	-.047	.133	.131	.084	-.247	.133	.131	14				
15	-.246	-.273	-.281	-.255	.139	.142	.110	.035	.133	.123	.084	-.059	.133	.123	15				
16	-.192	-.246	-.154	-.164	.137	.134	.110	.015	.143	.121	.108	-.007	.143	.121	16				
17	.288	.339	.241	.313	.281	.340	.234	.252	.287	.378	.346	.326	.287	.378	17				
18	.618	.531	.640	.815	.820	.518	.881	.998	.822	.547	.626	.873	.822	.547	18				
19	.695	.647	.728	.736	.856	.586	.796	.931	.871	.616	.601	.945	.871	.616	19				
20	.645	.634	.686	.690	.780	.598	.578	.765	.777	.618	.594	.945	.777	.618	20				
21	-.083	-.093	-.054	.078	.169	.164	.153	.172	.203	.195	.175	.851	.203	.195	21				
22	.218	.124	-.002	.131	.383	.328	.228	.212	.391	.339	.250	.851	.391	.339	22				
23	-.313	-.337	-.354	-.241	.021	.065	.031	-.039	.034	.047	.029	.215	.034	.047	23				
24	-.230	-.224	-.314	-.408	.060	.061	.052	-.044	.074	.052	.064	.104	.074	.052	24				
25	-.056	-.056	-.056	-.019	.132	.127	.114	.093	.143	.106	.084	.064	.143	.106	25				
26	-.031	-.026	-.033	.054	.123	.118	.117	.112	.128	.121	.123	.064	.128	.121	26				
27	.734	.693	.761	.743	.891	.612	.847	.942	.908	.853	.609	.965	.908	.853	27				
28	.570	.570	.621	.603	.891	.453	.525	.684	.807	.515	.495	.675	.807	.515	28				



TABLE XLI  
PRESSURE COEFFICIENTS  $\frac{\Delta p}{q_0}$  OBSERVED ON WING

$\delta_{f,30} = 49.5$   $z/D = 1.008$

$\delta_{f,55} = 59.4$

Tube number	$\alpha = 0.0$						$\alpha = 57.8$						$\alpha = 57.8$						
	n = 2915			n = 2915			n = 2915			n = 2085			n = 2085			n = 2085			
	Spanwise station		140.5	Spanwise station		140.5	Spanwise station		140.5	Spanwise station		110.0	118.0	126.0	Spanwise station		110.0	118.0	126.0
1	.220	.193	.628	.293	.802	.476	.200	.007	.758	.744	.062	.002							
2	.084	.252	-.152	-.381	.056	.129	-.441	-.649	.011	.084	-.511	-.607							
3	.071	-.066	-.247	-.193	.009	-.126	-.363	-.356	-.035	-.036	-.312	-.607							
4	.021	-.136	-.172	-.091	-.041	-.173	-.239	-.181	-.072	-.139	-.206	-.168							
5	-.102	-.204	-.173	-.124	-.011	-.090	-.171	-.132	-.038	-.069	-.160	-.146							
6	.131	-.056	.308	.492	.150	.157	.603	.757	.127	.139	.667	.732							
7	.333	.302	.315	.350	.366	.291	.457	.525	.257	.321	.492	.509							
8	.485	.408	.384	.442	.521	.361	.422	.563	.514	.417	.480	.557							
9	.470	.492	.457	.484	.521	.453	.422	.563	.514	.417	.480	.557							
10	.475	.488	.465	.488	.553	.497	.456	.593	.506	.494	.514	.554							
11	.495	.444	.419	.497	.551	.481	.362	.529	.511	.518	.381	.698							
12	-.336	-.443	-.414	-.328	-.009	-.083	.500	.833	.502	.499	.797	.823							
13	-.1273	-.1456	-.1453	-.102	-.051	-.199	-.420	.352	-.016	-.089	-.417	-.391							
14	-.973	-.1047	-.1181	-.902	-.051	-.199	-.441	-.734	-.110	-.206	-.557	-.900							
15	-.500	-.397	-.507	-.827	.171	.095	.016	-.693	.136	.086	-.259	-.281							
16	-.211	-.260	-.126	-.109	.176	.105	.050	-.017	.141	.100	-.052	-.067							
17	.351	.350	.317	.384	.440	.403	.058	-.021	.149	.120	-.052	-.067							
18	.624	.536	.842	.789	.629	.490	.366	.531	.504	.454	.441	.463							
19	.682	.619	.762	.760	.674	.530	.661	.859	.598	.542	.951	.920							
20	.649	.612	.694	.693	.674	.572	.838	.926	.605	.562	.912	.927							
21	.262	.253	.222	.321	.767	.572	.619	.782	.875	.614	.850	.828							
22	-.091	-.191	-.172	-.136	.549	.525	.340	.396	.434	.506	.422	.391							
23	-.626	-.620	-.587	-.400	.320	.117	-.049	-.050	-.042	.254	-.028	-.115							
24	-.282	-.343	-.443	-.342	-.051	-.023	-.082	-.144	-.194	-.028	-.086	-.285							
25	.007	.043	-.004	-.023	-.025	.002	-.091	-.091	-.110	.024	-.014	-.218							
26	.081	.093	.086	.045	.125	.092	.097	.076	.076	.120	.110	.055							
27	.742	.673	.797	.814	.775	.595	.110	.098	.112	.132	.115	.096							
28	.588	.631	.717	.703	.775	.685	.685	.819	.909	.473	.641	.898							

TABLE XLII  
PRESSURE COEFFICIENTS  $\frac{\Delta P}{q_0}$  OBSERVED ON WING

$\delta_{r,55} = 69.3$        $\delta_{r,30} = 28.5$        $z/d = 1.008$

Tube number	$\alpha = 0.0$				$\alpha = 56.8$				$\alpha = 98.2$				Tube number		
	Spanwise station				Spanwise station				Spanwise station						
	92.0	110.0	118.0	126.0	140.5	92.0	110.0	118.0	126.0	140.5	92.0	110.0		118.0	126.0
1	.236	.175	.742	.465	.627	.714	.096	.171	.357	.127	.553	.801	.570	.334	1
2	.150	.286	-.003	-.333	.096	.096	.171	.357	-.359	-.621	.178	.225	-.287	-.580	2
3	.179	-.085	-.229	-.205	-.097	.093	-.097	-.322	-.322	-.361	.220	-.084	-.309	-.359	3
4	.104	-.137	-.173	-.107	.028	.028	-.138	-.198	-.198	-.180	.121	-.124	-.188	-.196	4
5	-.028	-.137	-.145	-.116	.024	.024	-.044	-.118	-.118	-.104	.049	-.022	-.109	-.129	5
6	.487	-.171	.164	.457	.081	.204	.081	.567	.567	.732	.300	.012	.540	.692	6
7	.302	.271	.269	.352	.309	.351	.309	.446	.446	.718	.364	.322	.436	.510	7
8	.439	.450	.386	.440	.368	.497	.368	.430	.430	.559	.481	.451	.446	.558	8
9	.481	.517	.474	.487	.476	.506	.476	.492	.492	.588	.520	.563	.533	.597	9
10	.478	.529	.465	.517	.493	.545	.493	.497	.497	.608	.567	.570	.506	.533	10
11	.488	.510	.466	.532	.485	.563	.485	.667	.667	.859	.622	.547	.472	.448	11
12	-.829	-.749	-.944	-.852	-.312	-.414	-.312	-.727	-.727	-.849	-.518	-.309	-.917	-.844	12
13	-.700	-.647	-.1,196	-.989	.061	-.007	.061	-.201	-.201	-.482	-.126	.074	-.647	-.724	13
14	-.203	-.243	-.415	-.652	.145	.134	.145	.074	.074	-.076	.091	.143	-.111	-.203	14
15	-.159	-.210	-.282	-.225	.133	.134	.133	.074	.074	-.008	.096	.143	-.104	-.081	15
16	-.139	-.188	-.146	-.134	.148	.148	.148	.072	.072	.000	.101	.136	.086	-.037	16
17	.241	.289	.222	.294	.249	.249	.332	.288	.288	.248	.255	.357	.406	.260	17
18	.684	.561	.627	.812	.800	.800	.548	.624	.624	.847	.833	.645	.627	.338	18
19	.768	.669	.785	.800	.870	.870	.619	.823	.823	.959	.885	.734	.655	.840	19
20	.699	.670	.722	.724	.800	.800	.632	.615	.615	.769	.828	.749	.669	.803	20
21	-.054	-.042	-.046	.092	.149	.149	.144	.144	.144	.195	.168	.196	.181	.166	21
22	.241	.180	.074	.128	.394	.394	.335	.195	.195	.202	.429	.471	.297	.208	22
23	-.271	-.289	-.340	-.205	.032	.034	.032	-.008	-.008	-.070	-.017	.032	-.014	-.146	23
24	-.223	-.235	-.386	-.354	.019	.050	.019	.012	.012	-.079	.040	-.007	-.010	-.089	24
25	-.078	-.068	-.057	-.009	.109	.109	.085	.073	.073	-.058	.096	.084	-.067	-.049	25
26	-.052	-.038	-.019	.009	.103	.103	.085	.079	.080	-.086	.086	.089	.076	.074	26
27	.791	.713	.784	.802	.651	.890	.651	.619	.619	.898	.850	.776	.672	.984	27
28	.613	.613	.676	.678	.534	.890	.534	.527	.527	.702	.865	.627	.672	.863	28

TABLE XLIII  
PRESSURE COEFFICIENTS  $C_p$  OBSERVED ON WIND

$\theta_{r,55} = 69.3$        $\theta_{r,30} = 36.6$        $z/h = 1.00H$

Tube number	$\alpha = 0.0$			$\alpha = 58.2$			$\alpha = 58.8$		
	Spanwise station			Spanwise station			Spanwise station		
	92.0	110.0	126.0	92.0	110.0	126.0	92.0	110.0	126.0
1	.200	.195	.584	.595	.677	.659	.141	.096	.380
2	.279	.260	-.403	-.400	.132	.141	-.614	-.631	-.615
3	.274	.274	-.250	.163	.167	.167	-.371	-.380	-.351
4	.133	.138	-.119	-.129	.068	.068	-.189	-.200	-.205
5	-.026	-.016	-.115	-.119	.036	.039	-.127	-.132	-.126
6	.225	.234	.488	.459	.376	.374	.737	.744	.600
7	.325	.344	.342	.344	.378	.384	.361	.343	.487
8	.435	.440	.441	.449	.503	.507	.576	.571	.460
9	.453	.465	.528	.517	.525	.527	.505	.597	.509
10	.469	.483	.551	.529	.545	.550	.582	.583	.613
11	.480	.494	.446	.446	.634	.630	.555	.559	.675
12	.480	.480	.693	.667	.531	.503	-.573	-.629	.870
13	.625	.611	-.929	-.913	-.029	-.049	-.071	-.050	-.942
14	.642	.657	-1.087	-1.111	-.087	-.049	-.207	-.249	-.625
15	.144	-.178	-.741	-.748	.137	.143	.094	.189	-.106
16	.125	.125	-.224	-.229	.127	.142	.102	.091	-.064
17	.232	.242	-.145	-.145	.154	.149	.226	.246	-.032
18	.648	.623	.332	.305	.219	.223	.359	.359	.341
19	.748	.730	.882	.873	.902	.878	.942	.942	.867
20	.693	.694	.774	.756	.829	.806	.869	.870	.885
21	.067	.071	.084	.084	.846	.842	.881	.884	.788
22	.111	.120	-.154	-.154	.294	.295	.198	.200	.192
23	-.308	-.320	-.057	-.057	.313	.319	.304	.321	.091
24	-.234	-.241	-.477	-.477	-.020	-.019	-.184	-.184	-.175
25	-.073	-.073	-.084	-.084	-.002	-.004	-.037	-.041	-.101
26	-.048	-.048	.009	.009	.089	.084	.035	.027	-.056
27	.788	.784	.869	.863	.942	.924	.962	.962	.881
28	.644	.644	.777	.765	.482	.482	.514	.514	.509

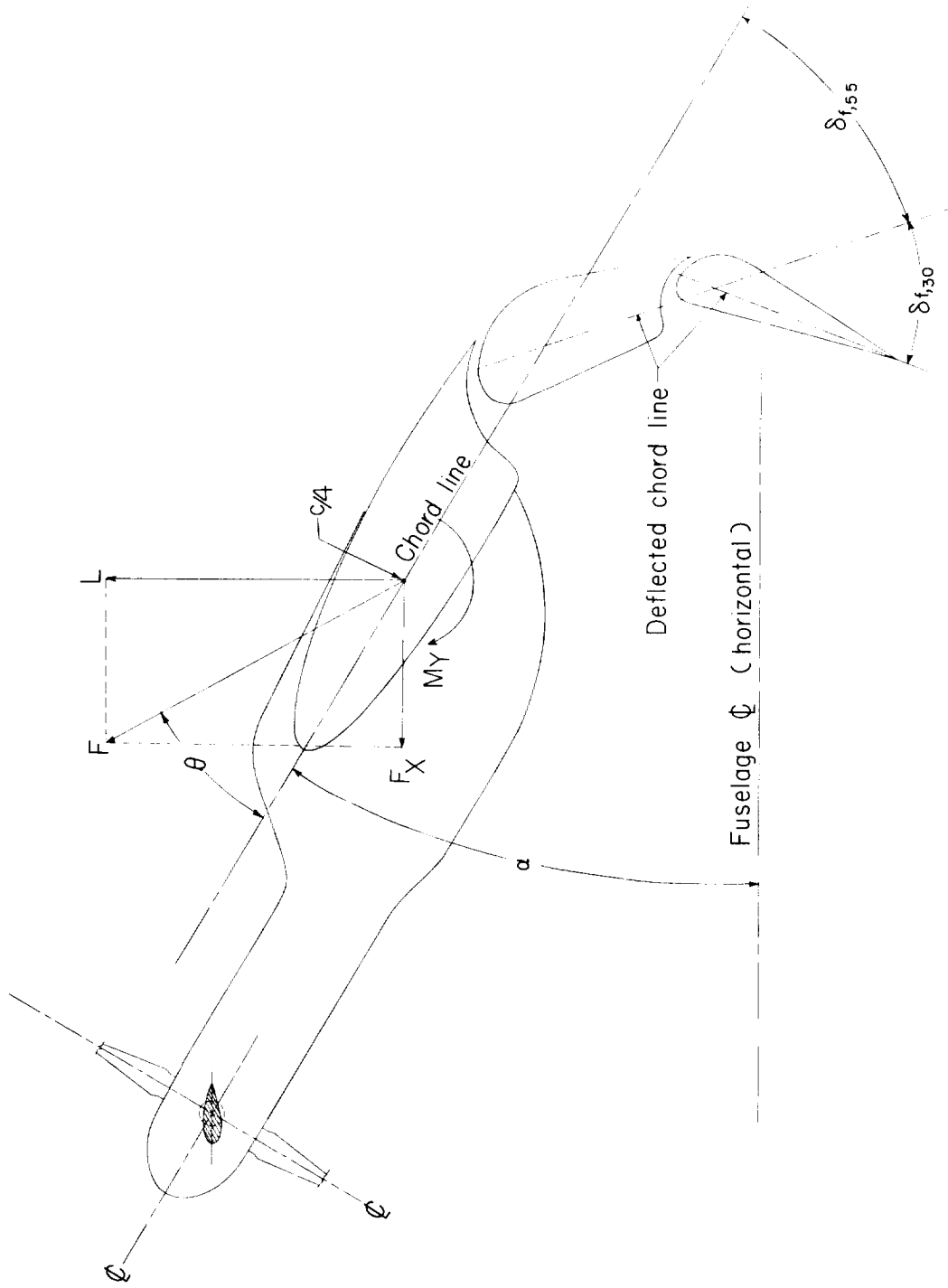


Figure 1.- Conventions used to define positive sense of forces, moments, and angles.

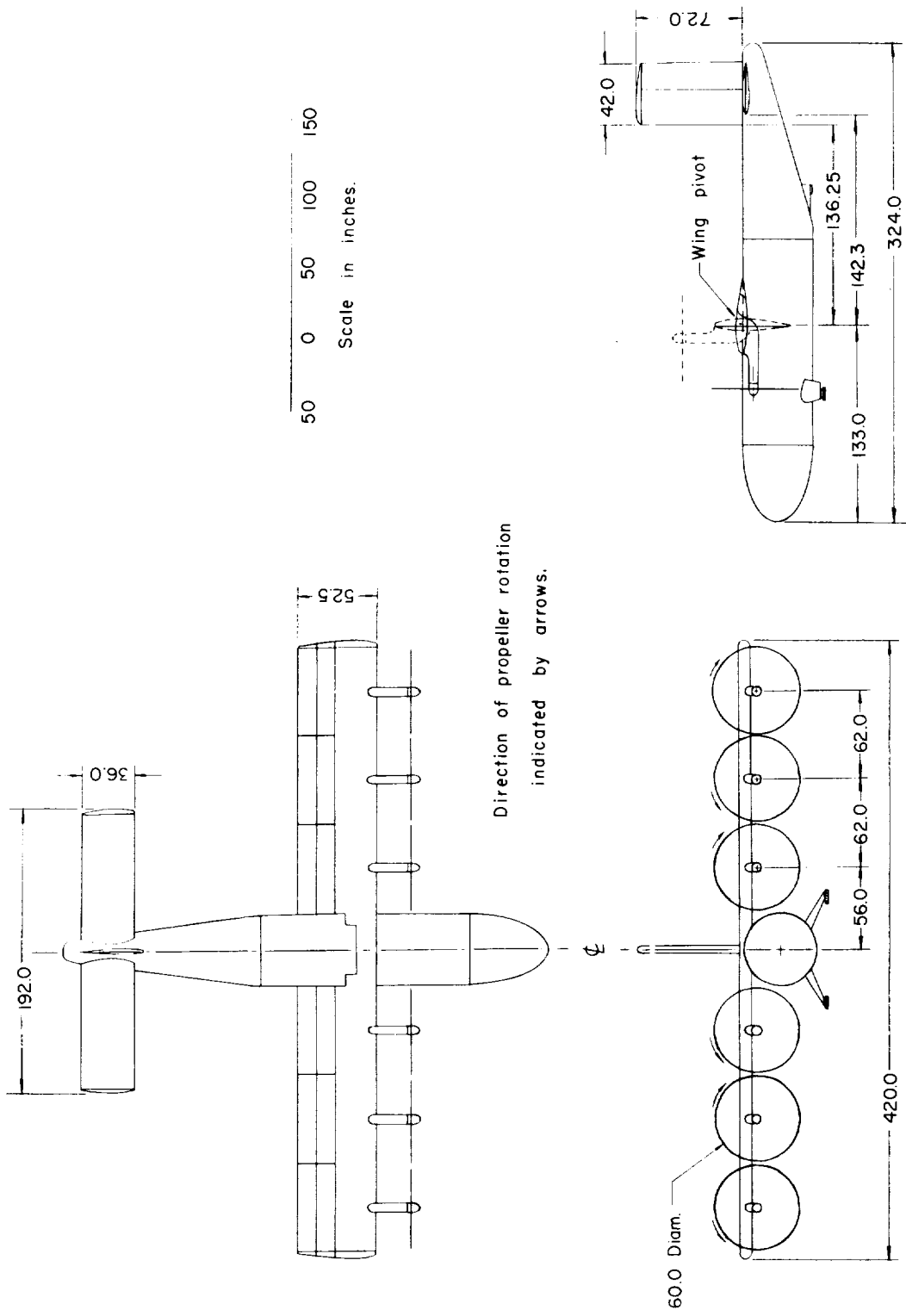
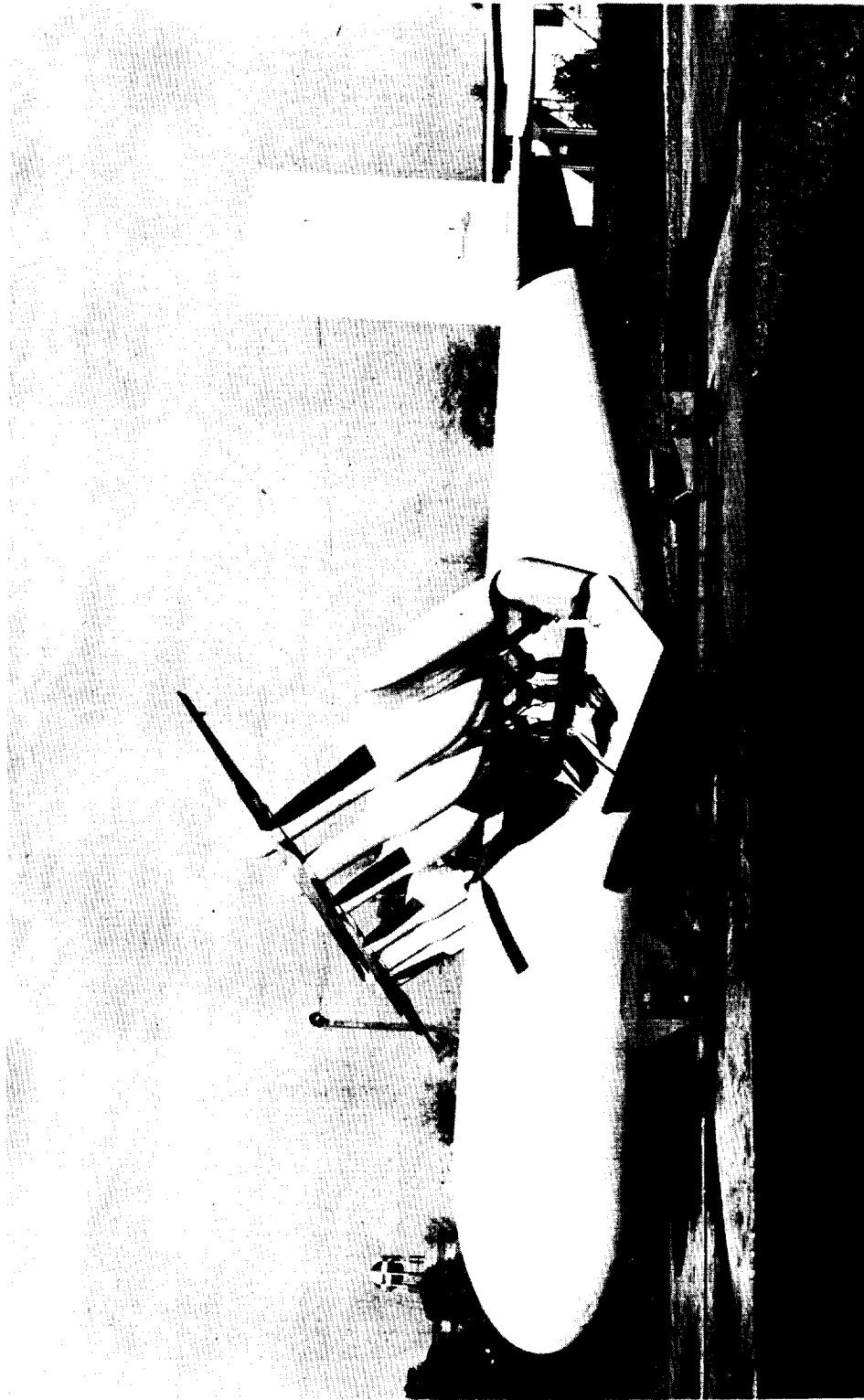


Figure 2.- Three-view sketch of model. All dimensions are in inches.





L-59-7989

Figure 3.- Model on balance at  $z/D = 1.008$ .

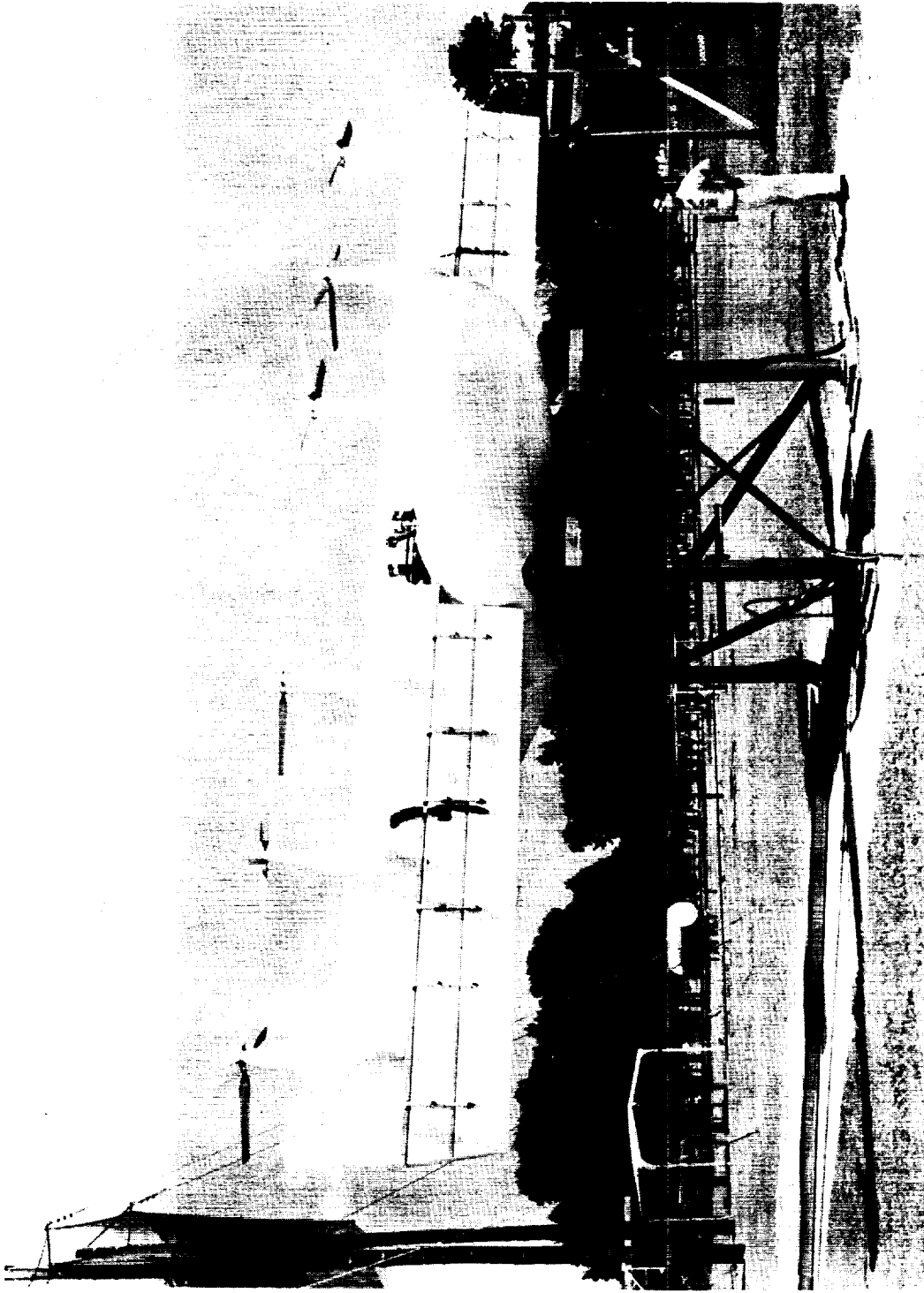


Figure 4.- Model on balance at  $z/D = 2.425$ . Wing vertical. L-59-4896

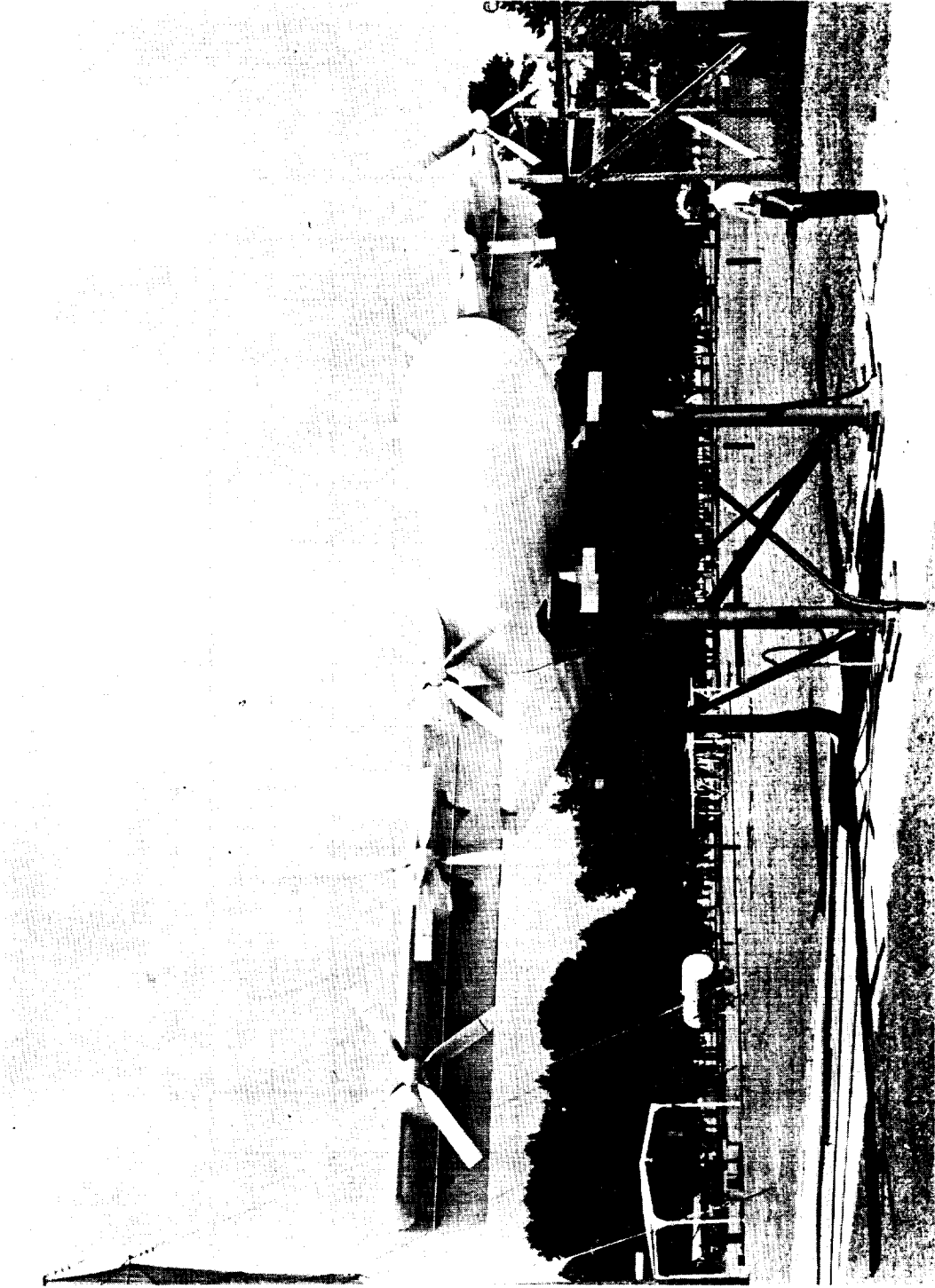


Figure 5.- Model on balance at  $z/D = 2.425$ . Wing horizontal. L-59-4897

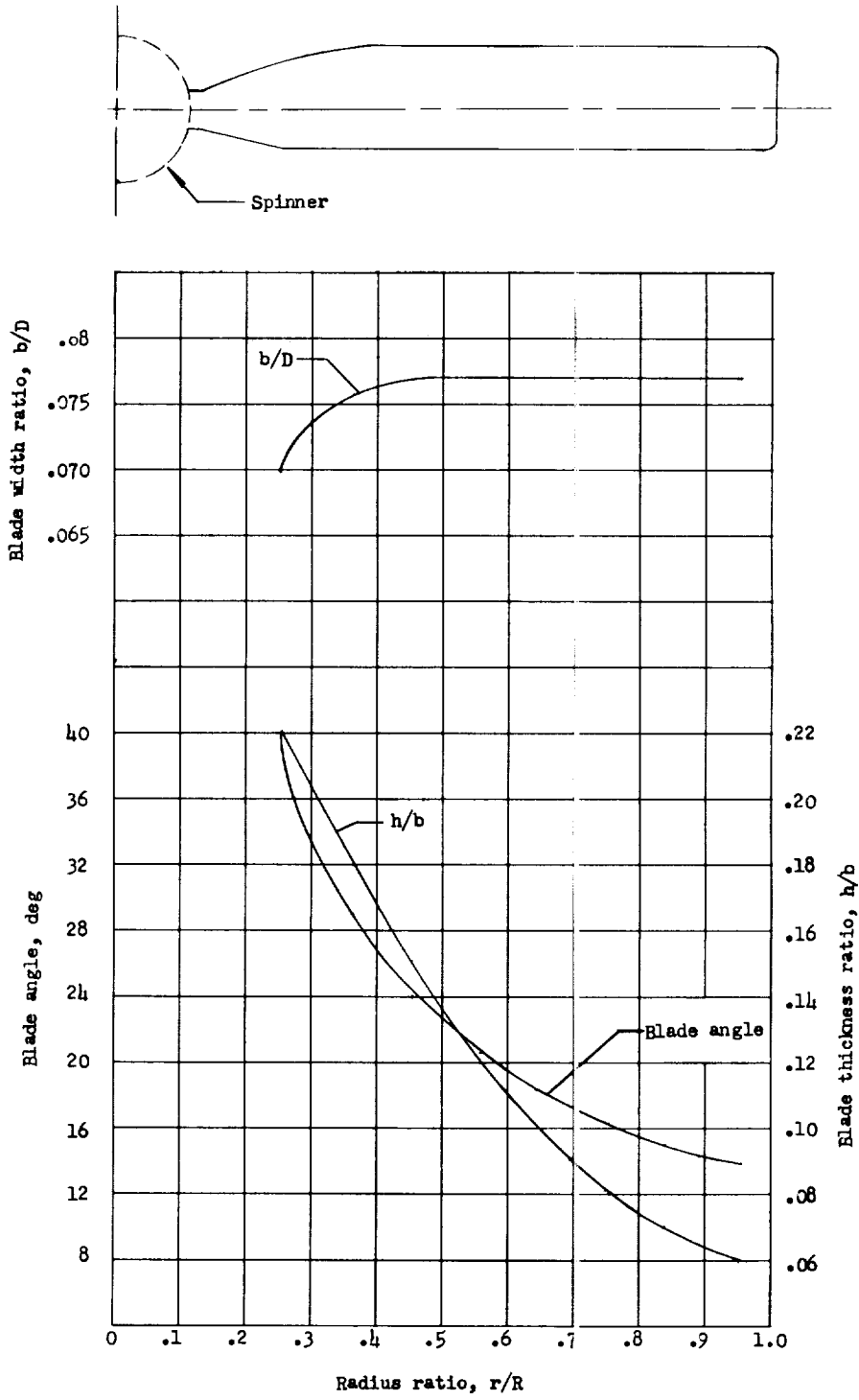


Figure 6.- Propeller blade form curves.

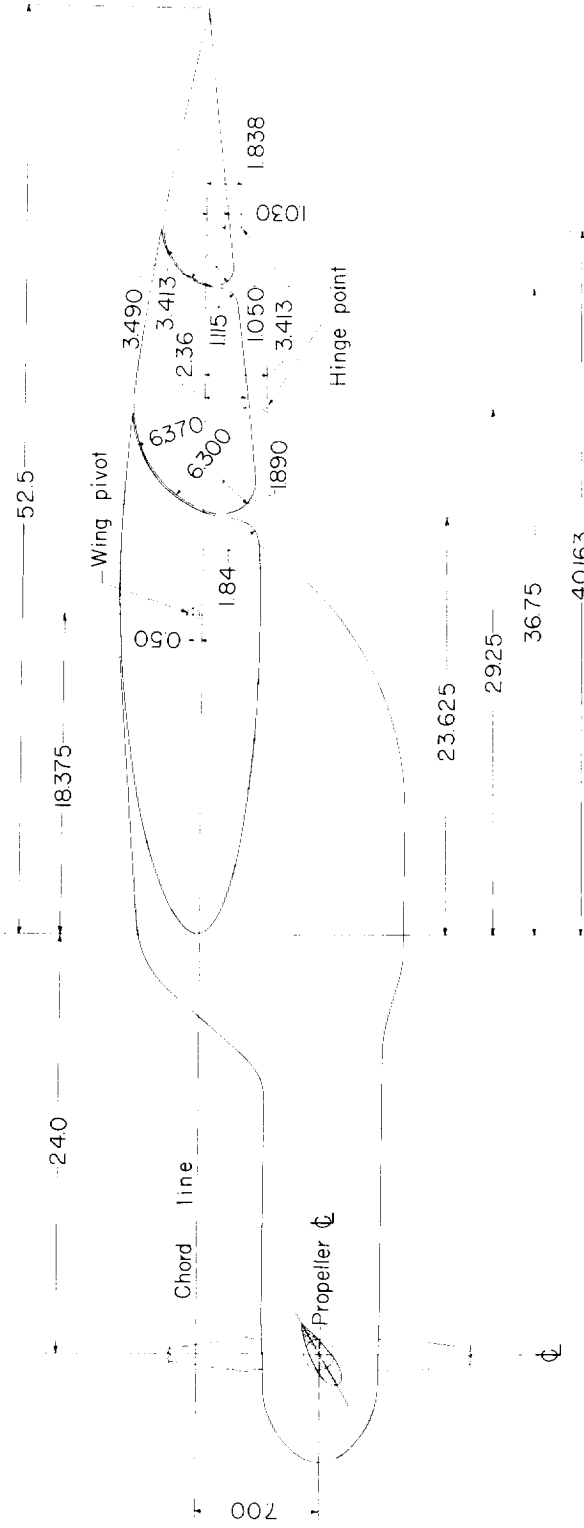


Figure 7.- Geometric characteristics of wing section. All dimensions are in inches.

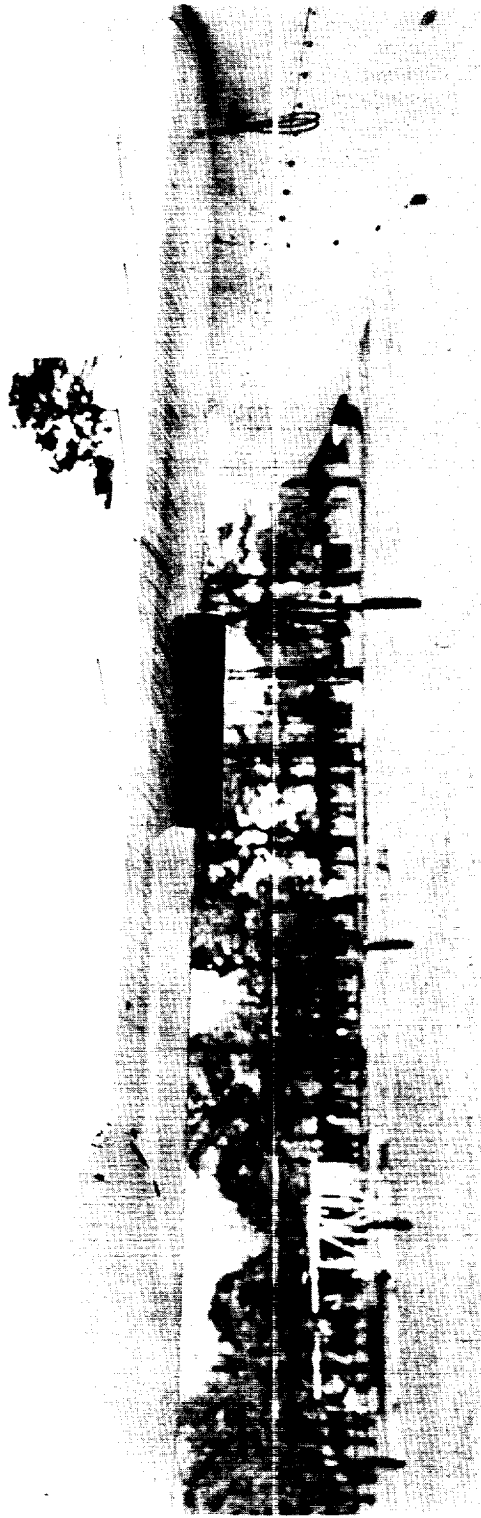


Figure 8.- Horizontal stabilizer.

L-59-7992

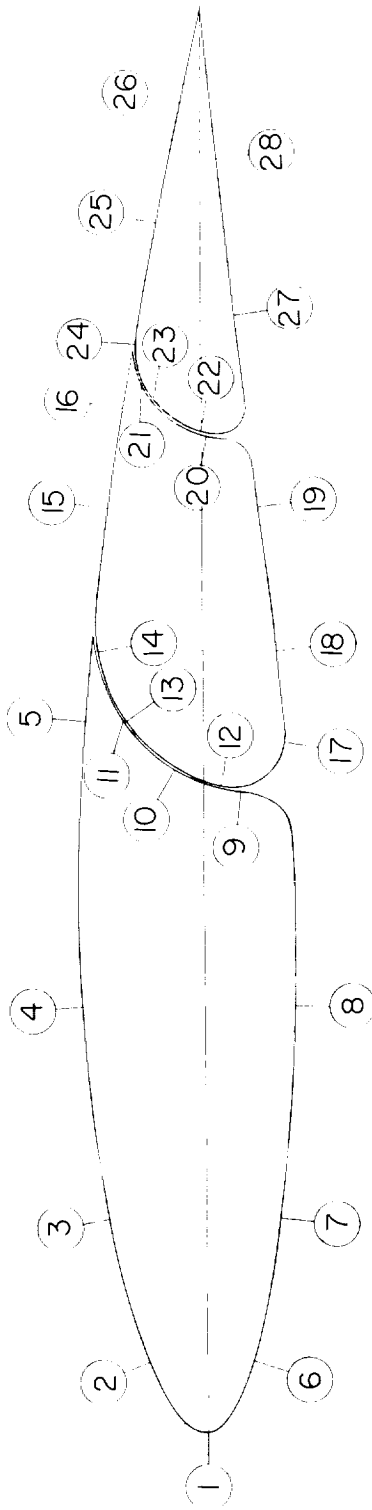
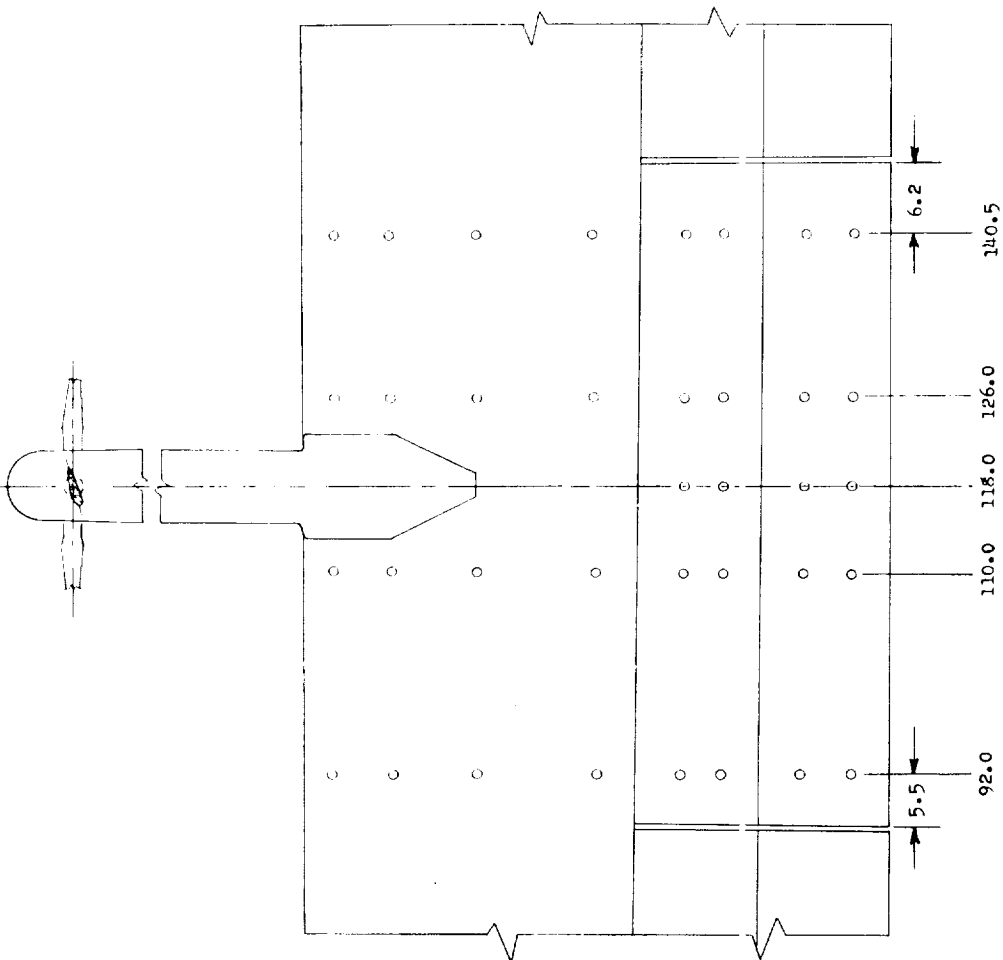


Figure 9.- Orifice locations on wing and flaps.

Propeller  $\xi$



Spanwise wing stations (inches from  $\xi$  of model)

Orifice Locations

Tube Number	Station, percent chord	Ordnate, percent chord
1	0	0
2	5.0	3.96
3	15.0	6.76
4	30.0	8.53
5	50.0	8.19
6	5.0	-3.18
7	15.0	-5.14
8	30.0	-6.25
9	45.0	-2.59
10	46.6	1.75
11	50.0	5.41
12	45.0	-2.03
13	50.0	5.26
14	55.0	7.33
15	65.0	6.42
16	71.5	5.41
17	48.3	-5.52
18	55.0	-5.05
19	65.0	-3.92
20	69.9	-0.90
21	73.4	0.00
22	70.0	-0.95
23	72.6	3.26
24	76.5	4.48
25	84.9	3.01
26	93.3	1.37
27	78.5	-2.19
28	89.5	-1.07

Figure 9.- Concluded.



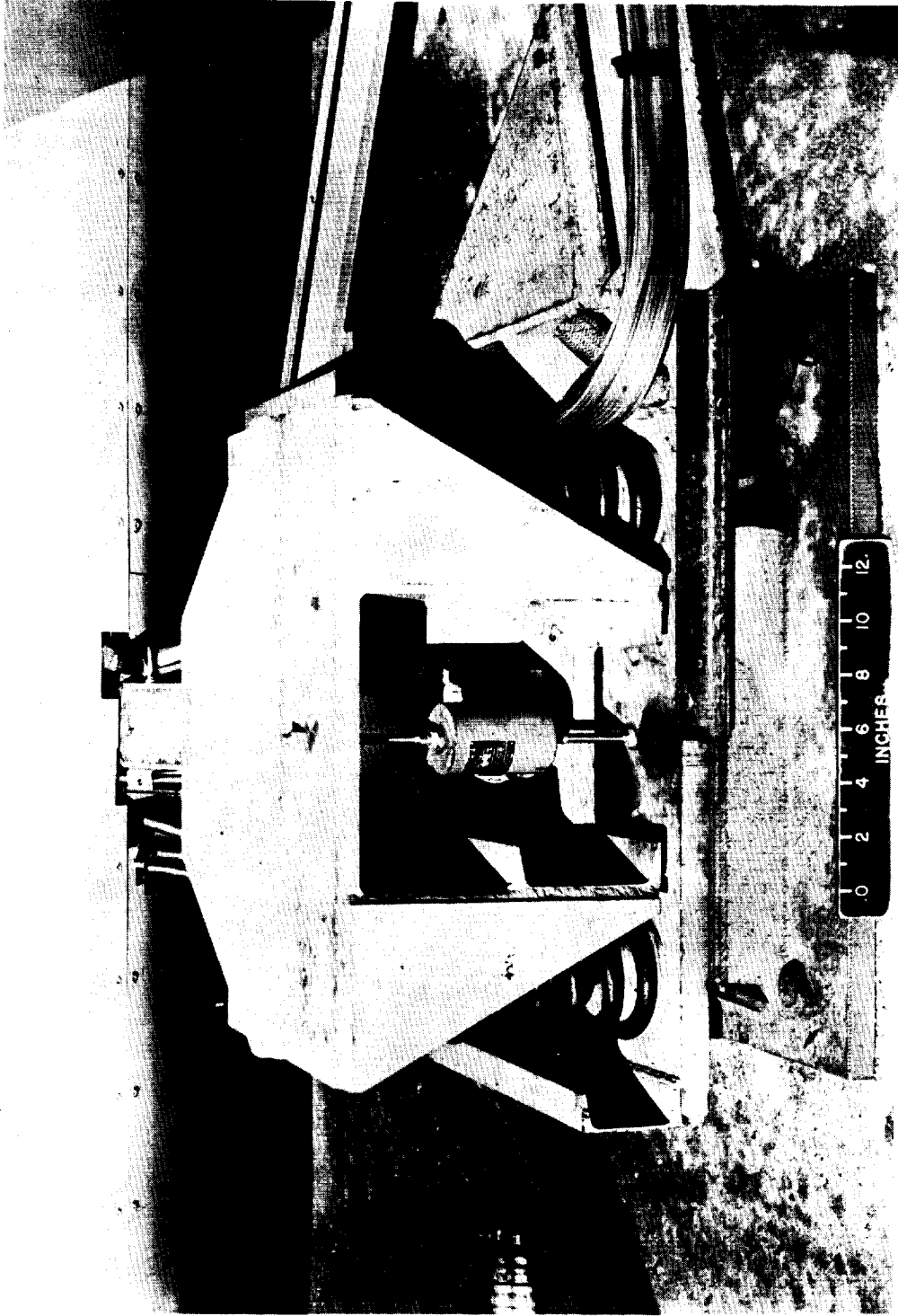


Figure 10.- Left front load cell support assembly. L-59-7994

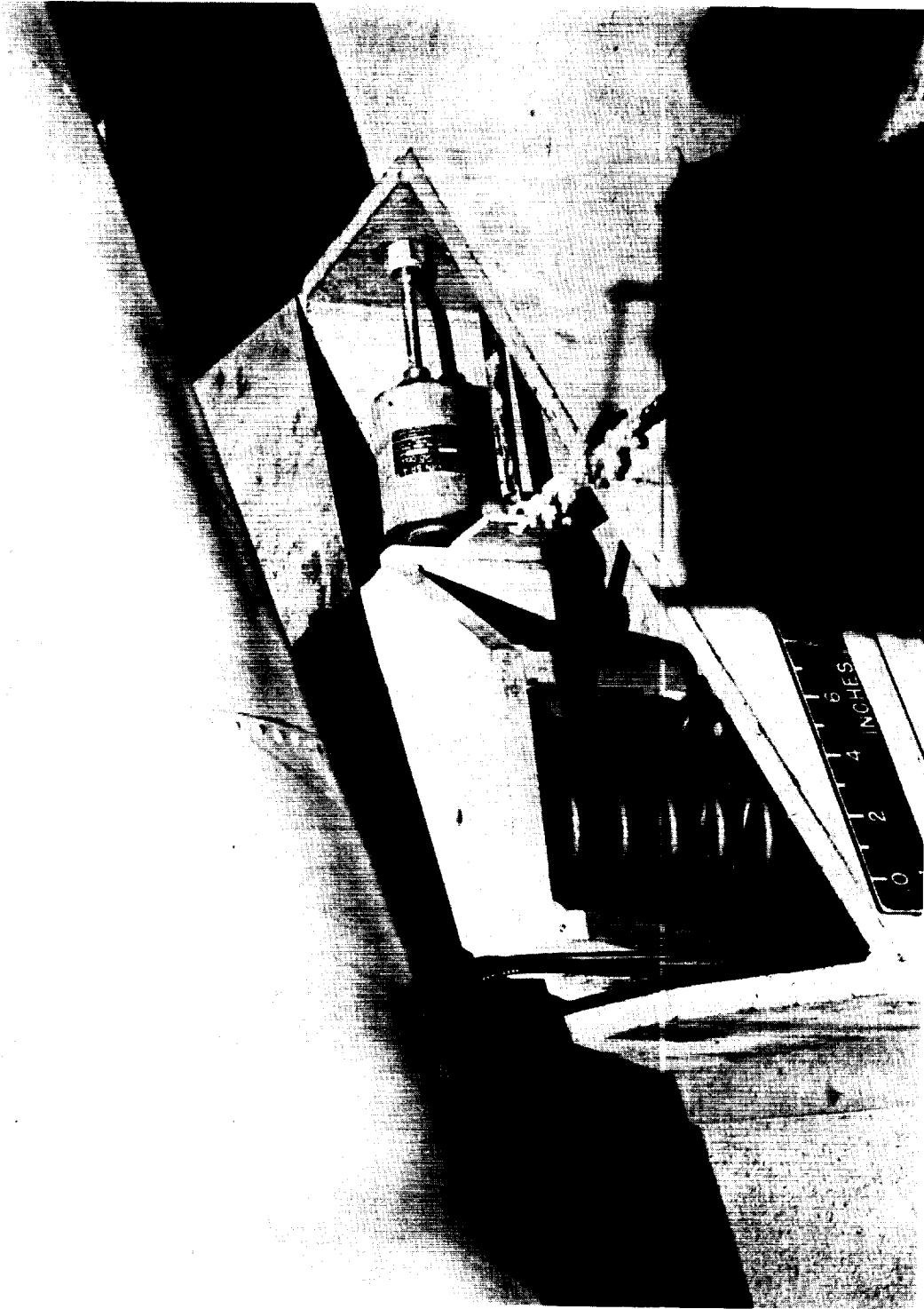


Figure 11.- Rear load cell support assembly. L-59-7995