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## MEMORANDUM

SOME EFFECTS OF HORIZONTAL-TAIL POSITION ON  
THE VERTICAL-TAIL PRESSURE DISTRIBUTIONS OF A COMPLETE  
MODEL IN SIDESLIP AT HIGH SUBSONIC SPEEDS

By William J. Alford, Jr.

Langley Research Center  
Langley Field, Va.

**NATIONAL AERONAUTICS AND  
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## SUMMARY

An investigation has been made in the Langley high-speed 7- by 10-foot tunnel of some effects of horizontal-tail position on the vertical-tail pressure distributions of a complete model in sideslip at high subsonic speeds. The wing of the model was swept back  $28.82^\circ$  at the quarter-chord line and had an aspect ratio of 3.50, a taper ratio of 0.067, and NACA 65A004 airfoil sections parallel to the model plane of symmetry. Tests were made with the horizontal tail off, on the wing-chord plane extended, and in T-tail arrangements in forward and rearward locations. The test Mach numbers ranged from 0.60 to 0.92, which corresponds to a Reynolds number range from approximately  $2.93 \times 10^6$  to  $3.69 \times 10^6$ , based on the wing mean aerodynamic chord. The sideslip angles varied from  $-3.9^\circ$  to  $12.7^\circ$  at several selected angles of attack.

The results indicated that, for a given angle of sideslip, increases in angle of attack caused reductions in the vertical-tail loads in the vicinity of the root chord and increases at the midspan and tip locations, with rearward movements in the local chordwise centers of pressure for the midspan locations and forward movements near the tip of the vertical tail. At the higher angles of attack all configurations investigated experienced outboard and rearward shifts in the center of pressure of the total vertical-tail load. Location of the horizontal tail on the wing-chord plane extended produced only small effects on the vertical-tail loads and centers of pressure. Locating the horizontal tail at the tip of the vertical tail in the forward position caused increases in the vertical-tail loads; this configuration, however, experienced considerable reduction in loads with increasing Mach number. Location of the horizontal tail at the tip of the vertical tail in the rearward position produced the largest increases in vertical-tail loads per degree sideslip angle; this configuration experienced the smallest variations of loads with Mach number of any of the configurations investigated.

## INTRODUCTION

The need for a better understanding of the nature of the air flow at the tail and its effects on the resulting vertical-tail loads has become increasingly important because several high-speed airplanes in current use have, in maneuvering flight, been subjected to extreme vertical-tail loads and experienced losses in directional stability. Of the many variables affecting the aerodynamic characteristics of a particular vertical-tail configuration, wing position and fuselage cross-sectional shape have been studied and the results of these investigations have been presented in references 1 to 5. A recent experimental investigation (ref. 6) has shown that the directional stability of a complete airplane model is also affected appreciably by the location of the horizontal tail.

The purpose of this paper is to present the results of an experimental investigation made to determine the effects of horizontal-tail position on the chordwise pressure distributions and integrated loads on the vertical tail of a complete model in sideslip at high subsonic speeds. The complete model consisted of a highly tapered wing (taper ratio, 0.067) of aspect ratio 3.50 with a quarter-chord sweepback of  $28.82^\circ$ , located on the center line of a fuselage which had an ogival nose, a cylindrical center section, and a truncated tail cone. The vertical-tail aspect ratio was 1.02 and the quarter-chord sweepback angle was  $28.35^\circ$ . The triangular horizontal tail was of aspect ratio 4.00 and was located either on the wing-chord plane or at the tip of the vertical tail in T-tail arrangements. Two chordwise locations were investigated for the T-tail configurations: one with the horizontal tail forward (zero trailing-edge overhang) and one with the horizontal tail rearward (zero leading-edge overhang). The sideslip-angle range extended from  $-3.9^\circ$  to  $12.7^\circ$  at angles of attack of approximately  $-9^\circ$ ,  $0^\circ$ ,  $9^\circ$ , and  $15^\circ$ . The test Mach numbers varied from 0.60 to 0.92 with the corresponding Reynolds number (based on wing mean aerodynamic chord) varying from  $2.93 \times 10^6$  to  $3.69 \times 10^6$ .

## SYMBOLS

$C_L$	lift coefficient, $\frac{\text{Lift}}{qS}$
$S$	wing area, sq ft
$C_p$	pressure coefficient, $\frac{p_l - p_\infty}{q}$

$p_l$	local static pressure, lb/sq ft
$p_\infty$	free-stream static pressure, lb/sq ft
$q$	dynamic pressure, $\frac{1}{2}\rho V_\infty^2$ , lb/sq ft
$\rho$	air density, slugs/cu ft
$V_\infty$	free-stream velocity, ft/sec
$M$	Mach number
$c_v$	local vertical-tail chord, ft
$\bar{c}_v$	vertical-tail mean aerodynamic chord, ft
$c_{v,av}$	average vertical-tail chord, ft
$\bar{c}$	wing mean aerodynamic chord, ft
$b_v$	exposed vertical-tail span, ft
$x$	chordwise distance from leading edge of vertical-tail local chord, ft
$z$	spanwise distance from vertical-tail root chord (vertical-tail root chord 0.154 ft above fuselage center line), ft
$z'$	spanwise distance from vertical-tail root chord to vertical-tail mean aerodynamic chord, ft
$\Delta\left(\frac{x}{c_v}\right)$	increment of vertical-tail local chord over which the pressure at a particular orifice is assumed to act (distance between points midway between adjacent orifices)
$c_{n,v}$	vertical-tail section normal-force coefficient, $\sum_{\frac{x}{c_v}=0}^1 (C_{p,L} - C_{p,R}) \Delta\left(\frac{x}{c_v}\right)$

$c_{m,v}$  vertical-tail section pitching-moment coefficient about 0.25 local vertical-tail chord,

$$\sum_{\frac{x}{c_v}=0}^1 (C_{p,L} - C_{p,R}) \left( \frac{x}{c_v} - 0.25 \right) \Delta \left( \frac{x}{c_v} \right)$$

$\frac{\bar{x}}{c_v}$  local chordwise location of vertical-tail center of pressure,  
 $-\left( \frac{c_{m,v}}{c_{n,v}} - 0.25 \right)$

$\frac{\bar{x}}{c_v}$  chordwise center-of-pressure location of total vertical-tail load,

$$0.25 - \frac{\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta \beta c_{v,av}} \left[ \left( \frac{z'}{b_v} - \frac{z}{b_v} \right) \frac{b_v}{c_v} \tan \Lambda + \left( 0.25 - \frac{\bar{x}}{c_v} \right) \frac{c_v}{c_v} \right] d \left( \frac{z}{b_v} \right)}{\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta \beta c_{v,av}} d \left( \frac{z}{b_v} \right)}$$

$\frac{\bar{z}}{b_v}$  spanwise center-of-pressure location of total vertical-tail load,

$$\frac{\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta \beta c_{v,av}} \frac{z}{b_v} d \left( \frac{z}{b_v} \right)}{\int_0^{1.0} \frac{\Delta c_n c_v}{\Delta \beta c_{v,av}} d \left( \frac{z}{b_v} \right)}$$

$\alpha$  angle of attack, deg

$\beta$  angle of sideslip, deg

$i_t$  horizontal-tail incidence angle, deg

$\Lambda$  sweep angle of quarter-chord line, deg

Subscripts:

R right side of vertical tail looking upstream

L left side of vertical tail looking upstream

## MODELS AND APPARATUS

Details of the complete model as tested are presented in figure 1 and a photograph of a typical model mounted on the sting-support system is presented as figure 2. The fuselage had a fineness ratio of 10.94 and was constructed of aluminum. The physical characteristics of the fuselage including afterbody ordinates are given in figure 1(d). The aluminum wing was swept back  $28.82^\circ$  at the quarter-chord line (wing 80-percent-chord line was unswept) and had an aspect ratio of 3.50, a taper ratio of 0.067, and NACA 65A004 airfoil sections parallel to the fuselage center line. The triangular horizontal tail was made of steel and covered with plastic and fiber glass; it had an aspect ratio of 4.00 and NACA 65A006 airfoil sections parallel to the fuselage center line. The horizontal tail was tested in three locations: on the wing chord line, and atop the vertical tail in both forward and rearward locations. These locations are shown in figure 1(b). The vertical tail was made of steel and was covered with plastic and fiber glass; it had an aspect ratio of 1.02 (based on exposed area and span), a taper ratio of 0.46, quarter-chord sweep of  $28.35^\circ$ , and NACA 65A006 airfoil sections. Chord-wise pressure orifices (in rows parallel to the fuselage center line) were located at four spanwise stations. Details of the orifice locations are given in figure 1(c) and other details of model geometry are given in table I. The model of this investigation is identical to the model of reference 6.

Tests were made on the sting-support system shown in figure 2. With this system the model can be remotely operated through an angle-of-attack range of approximately  $26^\circ$  in the plane of the vertical strut. By utilization of couplings in the sting behind the model, the model can be rolled  $90^\circ$  so that either angle of attack or angle of sideslip can be the remotely controlled variable. With the wings vertical, the couplings can be used to support the model at fixed angles of attack while the model is tested through the angle-of-sideslip range.

With the models at a given angle of attack and angle of sideslip, a record was taken of the pressures existing at the orifices by photographing a manometer board to which the orifices were connected. These pressures were reduced to coefficient form and are presented in tables II to VIII.

## TESTS AND CORRECTIONS

The tests were made in the Langley high-speed 7- by 10-foot tunnel through a Mach number range from 0.60 to 0.92, which corresponds to a Reynolds number range of approximately  $2.93 \times 10^6$  to  $3.69 \times 10^6$  based

on the wing mean aerodynamic chord. Pressure measurements on the vertical tail were made through a sideslip-angle range from  $-3.9^\circ$  to  $12.7^\circ$  at angles of attack of approximately  $-9^\circ$ ,  $0^\circ$ ,  $9^\circ$ , and  $15^\circ$  with the horizontal tail on the fuselage center line and at the tip of the vertical tail in T-tail arrangements. Two chordwise locations were investigated for the T-tail configurations: one with the horizontal tail forward (zero trailing-edge overhang) and one with the horizontal tail rearward (zero leading-edge overhang). (See fig. 1(b).) Pressure measurements were also made for configurations for which the wing, the horizontal tail, or both, were removed. The effects of changing the horizontal-tail incidence angle from  $0^\circ$  to  $-6^\circ$  were investigated for the T-tail configuration in the forward position.

Blockage corrections calculated by the method of reference 7 were applied to Mach number and dynamic pressure. Jet-boundary corrections calculated by the method of reference 8 have been applied to the angle of attack. Corrections for sting-support deflection have been applied to the angles of attack and sideslip.

#### PRESENTATION OF RESULTS

All the individual pressure coefficients are presented in tables II to VIII and some selected results from the investigation are presented in the following figures:

	Figure
Variation of model lift coefficient with angle of attack for horizontal tail off . . . . .	3
Effect of horizontal-tail position on vertical-tail pressure distribution . . . . .	4 and 5
Effect of sideslip angle on the spanwise variation of vertical-tail section normal-force coefficient . . . . .	6
Effect of horizontal-tail position on the spanwise variation of vertical-tail section normal-force coefficient per degree of sideslip . . . . .	7 and 8
Effect of horizontal-tail position and incidence on the spanwise variation of vertical-tail section normal-force coefficient per degree of sideslip . . . . .	9
Effect of horizontal-tail position and incidence on the chordwise location of vertical-tail local centers of pressure . . . . .	10
Effect of horizontal-tail position and incidence on the variation of the center of pressure of the total vertical-tail load with angle of attack . . . . .	11



## DISCUSSION

## Vertical-Tail Loads

For a given angle of attack, increasing the angle of sideslip caused increases in the vertical-tail section loads for all configurations investigated, except at the highest sideslip angle ( $12.7^\circ$ ) where in some cases load reductions occurred near the tip of the vertical tail (fig. 6).

For all configurations investigated (figs. 7 to 9) the effects of increasing the angle of attack for small angles of sideslip was to cause decreases in the vertical-tail loads in the vicinity of the root chord and to cause increases near the midspan locations. The loss in vertical-tail loads near the root and gain in the midspan region can be attributed to the large flow angularities generated by the fuselage vortices which become stronger and asymmetrically located with respect to the tail plane as the angle of attack is increased (refs. 2 and 9). The addition of the wing on the fuselage center line produced no significant effects on the vertical-tail loads except at the highest angle of attack (approx.  $15^\circ$ , figs. 7 and 8) where the wing causes an outboard shift of the spanwise center of load. This shift in load is presumed to be due to the wing downwash causing the body separation vortices to cross the vertical tail at a lower level. This effect has also been noted in reference 2.

The addition of the horizontal tail on the wing-chord plane extended (figs. 6 to 8) produced only small increases in the vertical-tail loads because of its small end-plate effect. Locating the horizontal tail at the tip of the vertical tail in the forward position (zero trailing-edge overhang, figs. 6 to 9) caused large increases in the vertical-tail loads, particularly near the tip of the vertical tail when compared with those for the horizontal-tail-off configuration or the configuration having the horizontal tail on the wing-chord plane extended. These increases in vertical-tail loads can be attributed to the more complete end-plate effectiveness of the horizontal tail on the vertical tail than to that of the horizontal-tail configuration located on the wing-chord plane extended. Moving the horizontal tail rearward on the vertical tail (zero leading-edge overhang, figs. 6 and 9) produced the largest vertical-tail loads of any of the configurations investigated; this indicated that the horizontal-tail configuration located rearward on the tip of the vertical tail produced a more favorable interference flow field. These results correlate well with the directional stability characteristics presented for the same configurations in reference 6. It should also be pointed out that the effects of the locations of the tip-mounted horizontal-tail configurations on the directional stability (ref. 6) are greater than the effects on the vertical-tail loads and, therefore, the actual loads

experienced by the vertical tail per unit restoring moment are less than for the horizontal tail off or horizontal tail located on wing-chord plane extended.

In general, the effects of change in Mach number on the vertical-tail loads for horizontal tail off and horizontal tail located on the wing-chord plane extended are small (figs. 6 to 8). For the T-tail configuration with the horizontal tail in the forward position, however, increasing the Mach number causes rather large reductions in the vertical-tail loads, particularly at zero angle of attack. The tail-load reductions are somewhat smaller at the higher angles of attack (figs. 7 and 8). The attendant reduction in directional stability for this configuration is noted in reference 6. The reductions in vertical-tail loads are presumed to be due to shock-interference effects between the horizontal and vertical tails. Reducing the horizontal-tail incidence for the horizontal-tail-forward T-tail configuration caused decreases in the vertical-tail loads, with these decreases becoming larger as the Mach number was increased. With the horizontal tail in the rearward position (figs. 6 and 9), the effects of increasing Mach number were to cause only small reductions in the vertical-tail loads, except at the highest Mach number where the loads again increase. These Mach number effects for the horizontal-tail-rearward T-tail configuration are presumed to be due to the shift in the position of the peak minimum pressures associated with the maximum thicknesses of the horizontal tail relative to the vertical-tail lifting pressures and, thereby, the severity of the shock interactions was reduced and more desirable load variations with Mach number were produced for this tail position than for the other horizontal-tail locations investigated.

#### Vertical-Tail Center of Pressure

In general, for the various horizontal-tail locations, the effects of increasing the angle of sideslip were to cause rearward movements in the local chordwise centers of pressure for the midspan locations and forward movements near the tip of the vertical tail (fig. 10). Increasing the angle of attack also caused rearward movements of the chordwise centers of pressure except at the tip of the vertical tail where some small forward movement was in evidence (fig. 10).

At small angles of attack and sideslip the addition of the horizontal tail tended to reduce the spanwise variations of the vertical-tail local chordwise centers of pressure. As the angles of attack and sideslip were increased, however, the effects of horizontal-tail position decreased and the local centers of pressure were essentially the same for all tail positions (fig. 10).

Increasing Mach number generally caused forward shifts in the vertical-tail chordwise centers of pressures for the lower spanwise locations and a rearward shift in the vicinity of the tip of the vertical tail. The largest Mach number effects were exhibited by the T-tail configuration with the horizontal tail mounted in the forward location and the smallest Mach number effects were exhibited by the T-tail arrangement with the horizontal tail mounted in the rearward location (fig. 10).

The center of pressure of the total vertical-tail load for the wing—fuselage—vertical-tail configuration (fig. 11) at a Mach number of 0.80 was located at approximately 45 percent of the vertical-tail span and 25 percent of the tail mean aerodynamic chord and was essentially invariant with increases in angle of attack to an angle of approximately  $10^\circ$  where further increases in angle caused an outboard and rather abrupt rearward movement.

The addition of the horizontal tail on the wing-chord plane extended produced below  $10^\circ$  angle of attack, only small changes in the spanwise location of the center of pressure (fig. 11(a)) although the chordwise center-of-pressure location (fig. 11(b)) moved rearward gradually with moderate increases in angle of attack. Above  $\alpha = 10^\circ$ , the chord-plane-extended horizontal-tail configuration produced the largest outboard and rearward shifts in center-of-pressure location of any of the configurations investigated. Locating the horizontal tail at the tip of the vertical tail in the forward position caused greater outboard and rearward shifts in the center-of-pressure locations than those of the horizontal-tail-off configuration throughout the moderate angle-of-attack range at a Mach number of 0.80. Locating the horizontal tail in the rear position at the tip of the vertical tail produced outboard and rearward shifts in the center-of-pressure locations, when compared with those of the horizontal-tail-off configuration that were essentially constant throughout the angle-of-attack range. The large outboard shifts in spanwise center-of-pressure location above  $10^\circ$  angle of attack were, as mentioned previously, due to the fuselage vortices which cause flow-angularity reductions at the inboard spanwise locations and increases for the outboard locations.

In general, the effects of increasing Mach number on the locations of the vertical-tail center of pressure were small for all configurations, except for the T-tail arrangement with the horizontal tail in the forward position where increasing Mach number caused a considerable inboard movement.

TABLE I

## PHYSICAL CHARACTERISTICS OF THE MODEL

Wing:	
Span, ft . . . . .	2.49
Root chord, ft . . . . .	1.33
Tip chord, ft . . . . .	0.089
Mean aerodynamic chord, ft . . . . .	0.89
Area, sq ft . . . . .	1.77
Aspect ratio . . . . .	3.50
Taper ratio . . . . .	0.067
Quarter-chord sweep, deg . . . . .	28.82
Airfoil section . . . . .	NACA 65A004
Horizontal tail:	
Span, ft . . . . .	1.162
Root chord, ft . . . . .	0.581
Tip chord, ft . . . . .	0
Mean aerodynamic chord, ft . . . . .	0.388
Area, sq ft . . . . .	0.337
Aspect ratio . . . . .	4.00
Quarter-chord sweep, deg . . . . .	36.85
Airfoil section . . . . .	NACA 65A006
Vertical tail:	
Span (measured from root chord), ft . . . . .	0.683
Root chord (located 0.154 ft above fuselage center line), ft . . . . .	0.912
Tip chord, ft . . . . .	0.420
Mean aerodynamic chord, ft . . . . .	0.696
Area, sq ft . . . . .	0.454
Aspect ratio . . . . .	1.02
Taper ratio . . . . .	0.46
Quarter-chord sweep, deg . . . . .	28.35
Airfoil section . . . . .	NACA 65A006

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL

(a)  $\alpha = -9.4^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.832	.558	.557	-.164	-1.054	.829	.576	.577	-.180	-.780
.025	-.100	-.522	-.530	-.982	-2.458	-.071	-.542	-.549	-1.299	-1.155
.075	-.121	-.342	-.340	-.601	-.992	-.127	-.396	-.400	-.840	-1.120
.150	-.162	-.296	-.301	-.441	-.672	-.141	-.305	-.306	-.496	-1.054
.250	-.123	-.203	-.207	-.279	-.396	-.150	-.257	-.257	-.361	-.964
.350	-.123	-.175	-.179	-.217	-.302	-.175	-.241	-.244	-.301	-.792
.450	-.137	-.164	-.166	-.185	-.251	-.189	-.219	-.223	-.244	-.585
.550	-.109	-.127	-.127	-.126	-.192	-.148	-.157	-.161	-.169	-.424
.650	-.068	-.073	-.079	-.071	-.141	-.102	-.105	-.106	-.107	-.309
.750	-.027	-.016	-.019	-.014	-.100	-.039	-.045	-.042	-.043	-.203
.850	.027	.037	.036	.039	-.033	.021	.025	.032	.032	-.097
.900	.048	.066	.066	.064	.008	.059	.062	.064	.062	-.040
	<i>Right side</i>									
.025	-.064	.276	.275	.539	.790	-.086	.340	.341	.596	.769
.075	-.150	.082	.082	.285	.498	-.143	.119	.123	.331	.512
.150	-.162	.002	.002	.151	.316	-.150	.018	.025	.183	.237
.250	-.127	-.027	-.028	.080	.204	-.164	-.041	-.040	.080	.199
.350	-.139	-.068	-.067	.011	.105	-.175	-.084	-.088	.009	.103
.450	-.148	-.105	-.104	-.041	.027	-.171	-.118	-.115	-.043	.018
.550	-.112	-.084	-.088	-.046	-.001	-.143	-.107	-.108	-.066	-.017
.650	-.066	-.055	-.053	-.030	-.001	-.102	-.075	-.074	-.037	-.015
.750	-.023	-.016	-.021	.002	.008	-.025	-.025	-.028	-.002	.002
.850	.043	.048	.043	.052	.050	.046	.057	.059	.059	.041
.900	.062	.062	.064	.066	.061	.055	.057	.057	.062	.034
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.829	.545	.550	-.016	-.273	.809	.515	.518	.062	-.173
.025	-.093	-.667	-.670	-.012	-.769	-.189	-.615	-.620	-.875	-.534
.075	-.139	-.440	-.441	-.938	-.752	-.157	-.373	-.374	-.854	-.552
.150	-.162	-.344	-.349	-.685	-.729	-.148	-.278	-.276	-.617	-.589
.250	-.166	-.273	-.276	-.425	-.700	-.096	-.200	-.200	-.375	-.594
.350	-.180	-.246	-.248	-.322	-.647					
.450	-.182	-.223	-.225	-.256	-.582					
.550	-.159	-.184	-.186	-.201	-.511					
.650	-.121	-.121	-.122	-.133	-.433	-.107	-.153	-.154	-.297	-.431
.750	-.036	-.052	-.051	-.066	-.352	-.059	-.121	-.115	-.345	-.394
.850	.002	.007	.004	-.007	-.279	-.025	-.086	-.081	-.375	-.355
.900	.023	.030	.027	.023	-.238	.023	-.045	-.044	-.370	-.336
	<i>Right side</i>									
.025	-.098	.362	.371	.614	.753	-.143	.264	.268	.470	.581
.075	-.159	.123	.126	.342	.503	-.175	.037	.032	.176	.291
.150	-.159	.034	.034	.192	.328	-.137	-.014	-.019	.066	.151
.250	-.180	-.043	-.044	.073	.188					
.350	-.189	-.098	-.097	-.002	.084					
.450	-.187	-.123	-.120	-.048	.011					
.550	-.159	-.118	-.115	-.071	-.021					
.650	-.109	-.086	-.081	-.043	-.031	-.089	-.071	-.072	-.043	-.035
.750	-.039	-.039	-.035	-.012	-.031	-.057	-.050	-.051	-.037	-.072
.850	.021	.041	.045	.052	-.015	.000	.002	.004	.007	-.063
.900	.050	.052	.057	.055	-.031	.014	.014	.011	.004	-.090

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(b)  $\alpha = -9.6^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.851	.879	.718	.352	-.129	.655	.862	.735	-.357	-.147
.025	.238	-.104	-.541	-.891	-1.541	.322	-.075	-.918	-1.640	-1.132
.075	.103	-.128	-.375	-.804	-1.223	.123	-.147	-.430	-1.419	-1.100
.150	-.010	-.188	-.343	-.513	-.964	.022	-.168	-.342	-.402	-.978
.250	-.037	-.147	-.224	-.295	-.573	-.051	-.180	-.287	-.318	-.849
.350	-.067	-.142	-.195	-.223	-.381	-.117	-.207	-.271	-.284	-.688
.450	-.117	-.162	-.181	-.188	-.292	-.172	-.227	-.244	-.243	-.542
.550	-.103	-.128	-.131	-.132	-.223	-.143	-.166	-.174	-.164	-.431
.650	-.074	-.071	-.071	-.065	-.168	-.100	-.115	-.105	-.092	-.344
.750	-.030	-.019	-.010	-.004	-.092	-.039	-.037	-.027	-.019	-.251
.850	.034	.040	.054	.053	-.017	.037	.038	.053	.058	-.153
.900	.062	.075	.091	.085	.019	.076	.079	.094	.093	-.097
	<i>Right side</i>									
.025	-.420	-.028	.285	.535	.764	-.894	-.075	.340	.581	.757
.075	-.411	-.148	.083	.282	.480	-.449	-.154	.115	.317	.500
.150	-.346	-.175	-.002	.143	.305	-.355	-.171	.019	.170	.325
.250	-.242	-.138	-.025	.073	.198	-.285	-.182	-.054	.064	.186
.350	-.213	-.156	-.077	-.002	.091	-.261	-.197	-.112	-.017	.082
.450	-.193	-.171	-.125	-.065	-.004	-.233	-.201	-.148	-.083	-.014
.550	-.131	-.125	-.106	-.074	-.037	-.178	-.165	-.134	-.101	-.055
.650	-.063	-.071	-.065	-.048	-.036	-.097	-.107	-.094	-.066	-.051
.750	-.007	-.016	-.019	-.019	-.028	-.013	-.021	-.019	-.014	-.031
.850	.067	.055	.053	.044	.021	.070	.061	.067	.054	.009
.900	.100	.081	.076	.068	.038	.097	.070	.074	.067	-.004
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.637	.870	.721	.346	-.040	.709	.849	.667	.317	.013
.025	.345	-.107	-1.097	-1.710	-.830	.233	-.210	-1.094	-.990	-.576
.075	.143	-.163	-.466	-1.495	-.849	.031	-.198	-.427	-.910	-.591
.150	.016	-.195	-.395	-.828	-.814	-.051	-.172	-.296	-.736	-.614
.250	-.065	-.203	-.305	-.355	-.718	-.024	-.116	-.218	-.488	-.597
.350	-.128	-.216	-.273	-.295	-.663					
.450	-.163	-.221	-.245	-.240	-.588					
.550	-.152	-.185	-.193	-.182	-.497					
.650	-.106	-.141	-.117	-.110	-.408	-.096	-.116	-.158	-.323	-.419
.750	-.051	-.030	-.036	-.040	-.333	-.044	-.051	-.112	-.359	-.378
.850	.022	.019	.028	.018	-.255	-.008	-.006	-.082	-.422	-.336
.900	.047	.049	.063	.047	-.225	.010	.016	-.045	-.407	-.320
	<i>Right side</i>									
.025	-1.122	-.087	.371	.601	.760	-1.082	-.156	.261	.462	.598
.075	-.489	-.174	.128	.326	.500	-.438	-.207	.004	.140	.290
.150	-.372	-.180	.025	.177	.329	-.276	-.159	-.039	.027	.138
.250	-.314	-.204	-.067	.048	.178					
.350	-.288	-.218	-.129	-.043	.056					
.450	-.247	-.213	-.161	-.095	-.022					
.550	-.189	-.185	-.148	-.112	-.062					
.650	-.111	-.116	-.100	-.071	-.065	-.114	-.089	-.071	-.062	-.065
.750	-.033	-.033	-.028	-.030	-.057	-.129	-.042	-.044	-.043	-.083
.850	.054	.051	.056	.041	-.030	-.048	.028	.024	.010	-.055
.900	.089	.078	.079	.061	-.049	-.018	.040	.031	.007	-.084

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(c)  $\alpha = -9.7^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.872	.893	.747	.466	.047	.690	.873	.765	.490	.033
.025	.237	-.115	-.549	-.860	-1.405	.311	-.095	-1.014	-1.426	-1.021
.075	.101	-.142	-.435	-.801	-1.075	.119	-.176	-.409	-1.291	-1.014
.150	-.019	-.207	-.395	-.676	-.906	.015	-.195	-.392	-1.158	-.898
.250	-.049	-.168	-.255	-.346	-.635	-.066	-.213	-.326	-.305	-.824
.350	-.084	-.162	-.223	-.239	-.455	-.136	-.239	-.306	-.259	-.703
.450	-.136	-.182	-.212	-.192	-.352	-.195	-.261	-.275	-.223	-.567
.550	-.126	-.144	-.152	-.126	-.260	-.165	-.199	-.195	-.149	-.456
.650	-.089	-.090	-.075	-.061	-.200	-.120	-.128	-.121	-.085	-.377
.750	-.044	-.036	-.021	-.002	-.124	-.054	-.047	-.035	-.012	-.290
.850	.025	.040	.049	.059	-.040	.034	.035	.050	.065	-.198
.900	.059	.075	.086	.090	.000	.076	.081	.092	.102	-.138
	<i>Right side</i>									
.025	-.420	-.017	.283	.534	.764	-.953	-.077	.330	.565	.750
.075	-.450	-.151	.079	.274	.481	-.542	-.166	.105	.303	.494
.150	-.396	-.186	-.018	.199	.308	-.384	-.185	.003	.160	.322
.250	-.268	-.152	-.042	.066	.190	-.308	-.203	-.074	.048	.183
.350	-.234	-.173	-.102	-.017	.084	-.279	-.226	-.139	-.044	.069
.450	-.211	-.192	-.154	-.091	-.017	-.246	-.227	-.182	-.116	-.031
.550	-.140	-.149	-.138	-.097	-.053	-.185	-.190	-.172	-.129	-.081
.650	-.069	-.088	-.092	-.071	-.058	-.100	-.114	-.128	-.091	-.081
.750	-.010	-.036	-.045	-.037	-.050	-.013	-.027	-.029	-.032	-.058
.850	.069	.051	.038	.039	.003	.074	.058	.055	.050	-.016
.900	.095	.078	.066	.060	.014	.101	.078	.063	.062	-.031
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.671	.879	.758	.473	.092	.727	.859	.695	.416	.102
.025	.338	-.124	-1.134	-1.495	-.855	.199	-.259	-1.082	-1.463	-.590
.075	.136	-.189	-.855	-1.343	-.868	-.002	-.243	-.870	-1.398	-.595
.150	.004	-.225	-.442	-1.228	-.729	-.086	-.205	-.266	-.897	-.611
.250	-.090	-.242	-.323	-.757	-.686	-.039	-.131	-.238	-.459	-.597
.350	-.154	-.249	-.301	-.185	-.652					
.450	-.194	-.249	-.278	-.164	-.604					
.550	-.178	-.212	-.216	-.155	-.534					
.650	-.131	-.156	-.136	-.102	-.453	-.113	-.129	-.189	-.385	-.418
.750	-.052	-.044	-.051	-.038	-.369	-.059	-.065	-.144	-.407	-.381
.850	.021	.017	.020	.022	-.291	-.016	-.014	-.122	-.472	-.341
.900	.044	.048	.055	.058	-.261	.000	.013	-.087	-.453	-.325
	<i>Right side</i>									
.025	-1.106	-.088	.360	.587	.754	-1.099	-.183	.234	.437	.595
.075	-.820	-.185	.112	.314	.496	-.730	-.250	-.045	.105	.275
.150	-.377	-.199	.006	.162	.324	-.239	-.188	-.079	-.022	.119
.250	-.320	-.236	-.102	.029	.168					
.350	-.299	-.247	-.168	-.074	.037					
.450	-.262	-.253	-.205	-.134	-.053					
.550	-.200	-.212	-.186	-.138	-.095					
.650	-.116	-.125	-.136	-.094	-.091	-.160	-.107	-.095	-.077	-.090
.750	-.035	-.040	-.041	-.039	-.082	-.164	-.057	-.062	-.055	-.102
.850	.058	.051	.042	.043	-.050	-.091	.021	.009	.009	-.071
.900	.089	.082	.070	.065	-.075	-.069	.033	.020	.006	-.101

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(d)  $\alpha = -9.8^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.885	.917	.801	.564	.167	.725	.887	.816	.588	.158
.025	.203	-.105	-.479	-.811	-1.258	.274	-.089	-.898	-1.226	-.855
.075	.067	-.137	-.464	-.746	-.985	.078	-.183	-.681	-1.133	-.858
.150	-.056	-.218	-.435	-.689	-.854	-.027	-.196	-.487	-1.067	-.833
.250	-.091	-.182	-.314	-.653	-.609	-.107	-.226	-.353	-.930	-.779
.350	-.125	-.180	-.252	-.352	-.529	-.205	-.275	-.376	-.522	-.668
.450	-.190	-.216	-.242	-.249	-.458	-.284	-.321	-.286	-.197	-.569
.550	-.186	-.174	-.154	-.119	-.362	-.244	-.227	-.187	-.099	-.469
.650	-.141	-.100	-.079	-.046	-.287	-.178	-.136	-.112	-.044	-.416
.750	-.080	-.036	-.011	.013	-.218	-.086	-.047	-.028	.015	-.361
.850	-.005	.044	.061	.072	-.133	.005	.044	.057	.083	-.289
.900	.038	.080	.095	.096	-.085	.055	.088	.102	.123	-.235
	<i>Right side</i>									
.025	-.390	-.024	.284	.529	.778	-.871	-.088	.312	.551	.752
.075	-.515	-.167	.072	.274	.496	-.714	-.192	.089	.292	.498
.150	-.476	-.214	-.026	.131	.316	-.576	-.210	-.012	.149	.325
.250	-.356	-.170	-.053	.055	.198	-.351	-.228	-.093	.029	.178
.350	-.289	-.198	-.122	-.032	.076	-.381	-.272	-.177	-.070	.056
.450	-.260	-.226	-.184	-.118	-.037	-.272	-.281	-.235	-.157	-.065
.550	-.169	-.174	-.176	-.137	-.088	-.204	-.228	-.222	-.182	-.124
.650	-.090	-.093	-.117	-.106	-.103	-.121	-.111	-.167	-.138	-.139
.750	-.024	-.031	-.053	-.060	-.108	-.027	-.028	-.020	-.052	-.115
.850	.052	.056	.039	.028	-.053	.056	.068	.055	.044	-.075
.900	.079	.087	.064	.056	-.049	.086	.086	.065	.060	-.097
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.701	.895	.811	.572	.197	.735	.869	.748	.510	.189
.025	.296	-.116	-.990	-1.291	-.920	.157	-.259	-.955	-1.280	-.592
.075	.095	-.196	-.854	-1.162	-.881	-.080	-.327	-.960	-1.237	-.589
.150	-.040	-.248	-.738	-1.130	-.583	-.192	-.251	-.735	-1.029	-.593
.250	-.145	-.286	-.453	-1.058	-.569	-.095	-.129	-.278	-.945	-.560
.350	-.237	-.303	-.286	-.823	-.583					
.450	-.293	-.309	-.231	-.367	-.581					
.550	-.255	-.252	-.216	-.151	-.559					
.650	-.189	-.125	-.134	-.040	-.517	-.153	-.136	-.185	-.351	-.425
.750	-.074	-.048	-.043	.019	-.453	-.087	-.067	-.181	-.416	-.399
.850	-.009	.028	.028	.060	-.373	-.037	-.009	-.187	-.442	-.365
.900	.023	.059	.065	.082	-.342	-.017	.020	-.172	-.406	-.358
	<i>Right side</i>									
.025	-.977	-.099	.337	.569	.752	-.973	-.202	.206	.420	.590
.075	-.864	-.211	.093	.298	.491	-1.000	-.354	-.109	.080	.270
.150	-.724	-.230	-.016	.150	.320	-.733	-.243	-.164	-.060	.103
.250	-.477	-.294	-.136	.001	.158					
.350	-.296	-.306	-.226	-.115	.011					
.450	-.253	-.302	-.278	-.204	-.099					
.550	-.225	-.250	-.238	-.214	-.156					
.650	-.139	-.104	-.168	-.142	-.156	-.216	-.119	-.112	-.098	-.141
.750	-.053	-.041	-.049	-.059	-.144	-.260	-.061	-.070	-.066	-.152
.850	.042	.062	.048	.036	-.095	-.204	.026	.014	.004	-.105
.900	.078	.094	.073	.060	-.127	-.197	.042	.025	.004	-.133



TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(e)  $\alpha = -9.8^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.919	.929	.811	.580	.101	.755	.884	.819	.601	.092
.025	.230	-.109	-.469	-.851	-1.456	.295	-.098	-.864	-1.172	-1.039
.075	.091	-.147	-.492	-.749	-1.184	.093	-.203	-.733	-1.090	-1.049
.150	-.040	-.237	-.471	-.724	-1.057	-.018	-.239	-.588	-1.049	-1.050
.250	-.080	-.214	-.509	-.734	-.801	-.101	-.253	-.553	-.976	-.972
.350	-.120	-.210	-.302	-.510	-.712	-.194	-.305	-.424	-.762	-.839
.450	-.195	-.254	-.285	-.382	-.626	-.300	-.367	-.410	-.545	-.722
.550	-.240	-.261	-.188	-.193	-.535	-.329	-.358	-.166	-.100	-.646
.650	-.172	-.138	-.079	-.063	-.470	-.216	-.143	-.081	-.029	-.630
.750	-.080	-.044	-.005	.009	-.417	-.066	-.045	-.010	.026	-.587
.850	.008	.032	.061	.066	-.341	.026	.041	.064	.086	-.516
.900	.050	.071	.098	.095	-.280	.075	.089	.108	.122	-.436
	<i>Right side</i>									
.025	-.365	-.031	.295	.544	.743	-.822	-.105	.318	.551	.708
.075	-.522	-.186	.078	.283	.428	-.731	-.220	.093	.290	.422
.150	-.494	-.241	-.031	.135	.230	-.616	-.245	-.015	.148	.233
.250	-.507	-.197	-.058	.055	.095	-.563	-.241	-.087	.032	.074
.350	-.332	-.223	-.128	-.042	-.043	-.417	-.308	-.180	-.079	-.066
.450	-.316	-.274	-.201	-.141	-.178	-.423	-.370	-.279	-.199	-.211
.550	-.185	-.252	-.223	-.180	-.246	-.168	-.338	-.307	-.250	-.299
.650	-.070	-.122	-.155	-.162	-.280	-.074	-.120	-.201	-.210	-.336
.750	-.003	-.044	-.068	-.096	-.302	.004	-.028	-.035	-.077	-.305
.850	.073	.046	.030	.009	-.230	.083	.062	.053	.026	-.246
.900	.103	.080	.062	.040	-.224	.110	.085	.072	.051	-.270
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.732	.892	.819	.591	.143	.764	.864	.757	.528	.133
.025	.309	-.124	-.938	-1.229	-1.066	.167	-.266	-.896	-1.187	-.807
.075	.107	-.215	-.827	-1.115	-1.004	-.076	-.362	-.920	-1.178	-.755
.150	-.034	-.268	-.789	-1.087	-.721	-.216	-.328	-.760	-.993	-.757
.250	-.145	-.320	-.709	-1.069	-.712	-.114	-.179	-.648	-.888	-.716
.350	-.257	-.394	-.623	-1.008	-.722					
.450	-.348	-.397	-.273	-.637	-.723					
.550	-.363	-.345	-.104	-.389	-.715					
.650	-.295	-.132	-.085	-.195	-.693	-.119	-.141	-.151	-.605	-.594
.750	-.031	-.032	-.022	-.041	-.640	-.057	-.069	-.141	-.503	-.577
.850	.015	.020	.040	.051	-.575	-.013	-.013	-.176	-.422	-.542
.900	.044	.051	.071	.081	-.532	.004	.015	-.189	-.376	-.525
	<i>Right side</i>									
.025	-.906	-.116	.334	.557	.700	-.897	-.207	.206	.412	.524
.075	-.823	-.232	.092	.291	.416	-.944	-.419	-.124	.061	.163
.150	-.763	-.253	-.018	.143	.228	-.756	-.316	-.201	-.083	-.024
.250	-.730	-.328	-.145	-.012	.042					
.350	-.662	-.413	-.253	-.144	-.121					
.450	-.296	-.394	-.337	-.266	-.256					
.550	-.097	-.347	-.349	-.313	-.342					
.650	-.068	-.075	-.192	-.235	-.352	-.126	-.113	-.090	-.146	-.313
.750	-.014	-.036	-.032	-.077	-.332	-.149	-.068	-.053	-.090	-.331
.850	.070	.058	.054	.021	-.253	-.153	.015	.023	-.016	-.264
.900	.100	.091	.081	.048	-.286	-.174	.033	.032	-.023	-.290

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(†)  $\alpha = 0^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.615	.643	.345	-.487	-1.393	.311	.736	.324	-.450	-1.181
.025	.269	-.092	-.520	-1.130	-2.260	.329	-.090	-.625	-1.156	-1.137
.075	.133	-.111	-.369	-.683	-1.607	.149	-.129	-.430	-1.042	-1.107
.150	.036	-.136	-.304	-.478	-.683	.052	-.141	-.316	-.776	-1.051
.250	.024	-.097	-.200	-.293	-.416	-.013	-.150	-.262	-.422	-1.007
.350	-.015	-.104	-.169	-.231	-.339	-.068	-.162	-.246	-.307	-.865
.450	-.054	-.113	-.155	-.193	-.286	-.094	-.164	-.223	-.247	-.655
.550	-.052	-.101	-.113	-.137	-.290	-.084	-.139	-.160	-.184	-.490
.650	-.034	-.064	-.067	-.091	-.200	-.066	-.104	-.107	-.121	-.362
.750	-.001	-.016	-.009	-.042	-.116	-.020	-.029	-.039	-.051	-.239
.850	.040	.033	.035	.014	-.023	.040	.029	.024	.026	-.125
.900	.070	.068	.061	.042	.014	.073	.063	.061	.061	-.060
	<i>Right side</i>									
.025	-.442	-.029	.319	.609	.847	-.613	-.062	.377	.606	.745
.075	-.364	-.108	.131	.348	.563	-.408	-.120	.152	.362	.528
.150	-.290	-.127	.038	.208	.377	-.313	-.134	.054	.217	.373
.250	-.197	-.101	.010	.133	.268	-.244	-.143	-.009	.121	.242
.350	-.167	-.108	-.030	.066	.173	-.225	-.155	-.060	.042	.154
.450	-.151	-.127	-.074	.005	.093	-.121	-.101	-.074	-.051	-.025
.550	-.103	-.099	-.055	-.002	.059	-.158	-.134	-.086	-.023	.033
.650	-.052	-.060	-.039	.007	.054	-.096	-.090	-.060	-.011	.026
.750	-.006	-.023	-.023	.010	.042	-.027	-.029	-.013	.017	.028
.850	.052	.042	.035	.059	.070	.047	.042	.040	.061	.045
.900	1.099	1.102	1.104	1.102	1.109	.073	.061	.059	.066	.038
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.225	.713	.261	-.410	-.672	.415	.681	.228	-.214	-.472
.025	.343	-.122	-.830	-1.142	-.732	.211	-.215	-.739	-.748	-.509
.075	.158	-.145	-.483	-1.098	-.711	.036	-.159	-.379	-.753	-.523
.150	.043	-.159	-.367	-.927	-.697	-.050	-.150	-.279	-.685	-.579
.250	-.024	-.162	-.283	-.599	-.669	-.096	-.155	-.234	-.515	-.562
.350	-.077	-.169	-.244	-.382	-.641					
.450	-.105	-.173	-.223	-.270	-.604					
.550	-.105	-.155	-.183	-.205	-.562					
.650	-.068	-.106	-.118	-.137	-.513	-.087	-.085	-.116	-.317	-.386
.750	-.034	-.041	-.051	-.072	-.441	-.038	-.032	-.074	-.289	-.362
.850	.020	.015	.012	-.016	-.372	-.013	.010	-.046	-.254	-.337
.900	.043	.033	.031	.007	-.341	.003	.022	-.037	-.224	-.323
	<i>Right side</i>									
.025	-.682	-.074	.393	.595	.700	-.611	-.132	.261	.408	.498
.075	-.465	-.136	.152	.348	.491	-.355	-.162	.024	.135	.247
.150	-.318	-.141	.056	.210	.342	-.262	-.150	-.048	.014	.098
.250	-.271	-.157	-.027	.100	.212					
.350	-.244	-.169	-.081	.019	.117					
.450	-.207	-.166	-.104	-.028	.042					
.550	-.161	-.143	-.097	-.037	.010					
.650	-.105	-.097	-.069	-.025	-.011	-.098	-.078	-.081	-.079	-.107
.750	-.043	-.048	-.030	-.011	-.032	-.068	-.034	-.060	-.067	-.113
.850	.040	.031	.040	.042	.023	-.015	.019	-.011	-.028	-.109
.900	.066	.059	.061	.054	-.051	-.006	.031	.000	-.032	-.134

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(g)  $\alpha = 0^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>					<i>Right side</i>				
.000	.713	.667	.534	.177	-.326	.484	.773	.533	.044	-.579
.025	.284	-.068	-.527	-1.160	-1.538	.330	-.081	-.931	-1.345	-.961
.075	.144	-.094	-.361	-.890	-1.260	.154	-.129	-.440	-1.131	-.969
.150	.043	-.140	-.315	-.505	-.978	.056	-.144	-.330	-.871	-.925
.250	.023	-.100	-.201	-.302	-.639	-.018	-.155	-.269	-.486	-.848
.350	-.024	-.115	-.164	-.236	-.408	-.076	-.177	-.247	-.311	-.749
.450	-.075	-.134	-.152	-.197	-.325	-.116	-.187	-.219	-.253	-.631
.550	-.070	-.106	-.102	-.136	-.283	-.105	-.152	-.152	-.180	-.515
.650	-.050	-.065	-.053	-.082	-.187	-.085	-.108	-.090	-.113	-.422
.750	-.019	-.017	-.013	-.026	-.128	-.027	-.022	-.014	-.031	-.326
.850	.036	.048	.070	.034	-.051	.043	.046	.063	.050	-.223
.900	.072	.083	.095	.070	-.006	.085	.089	.097	.095	-.158
	<i>Left side</i>					<i>Right side</i>				
.025	-.401	.015	.345	.606	.828	-.931	-.037	.379	.603	.749
.075	-.379	-.089	.158	.358	.562	-.428	-.118	.172	.367	.537
.150	-.310	-.118	.066	.224	.390	-.327	-.128	.072	.230	.383
.250	-.209	-.098	.029	.147	.280	-.263	-.144	.004	.124	.251
.350	-.182	-.118	-.022	.072	.170	-.235	-.161	-.053	.046	.149
.450	-.163	-.134	-.068	.004	.084	-.128	-.100	-.067	-.046	-.028
.550	-.105	-.104	-.058	-.005	.054	-.160	-.143	-.090	-.037	.022
.650	-.050	-.055	-.034	-.002	.034	-.090	-.094	-.056	-.023	.005
.750	.005	-.015	-.008	.010	.016	-.007	-.009	.001	.017	.011
.850	.069	.058	.053	.069	.048	.074	.066	.066	.075	.023
.900	1.164	1.166	1.167	1.166	1.160	.095	.086	.087	.087	.006
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>					<i>Right side</i>				
.000	.422	.748	.473	.060	-.404	.501	.704	.395	.075	-.285
.025	.337	-.118	-1.061	-.919	-.695	.209	-.241	-1.001	-.620	-.453
.075	.155	-.152	-.555	-.882	-.678	.008	-.184	-.555	-.611	-.455
.150	.039	-.171	-.383	-.825	-.664	-.073	-.163	-.276	-.606	-.514
.250	-.038	-.175	-.290	-.746	-.633	-.119	-.171	-.232	-.533	-.520
.350	-.097	-.186	-.246	-.613	-.602					
.450	-.136	-.189	-.218	-.448	-.565					
.550	-.127	-.166	-.170	-.308	-.518					
.650	-.091	-.104	-.093	-.188	-.481	-.084	-.066	-.102	-.308	-.364
.750	-.039	-.023	-.030	-.088	-.429	-.032	-.011	-.051	-.276	-.342
.850	.043	.034	.040	-.014	-.371	.002	.037	-.024	-.237	-.314
.900	.050	.061	.067	.020	-.345	.023	.055	-.013	-.211	-.299
	<i>Left side</i>					<i>Right side</i>				
.025	-1.041	-.046	.395	.595	.715	-.958	-.134	.264	.415	.528
.075	-.497	-.138	.167	.352	.513	-.439	-.184	.007	.135	.268
.150	-.343	-.138	.073	.221	.359	-.264	-.163	-.064	-.002	.102
.250	-.298	-.171	-.021	.103	.228					
.350	-.257	-.181	-.081	.014	.109					
.450	-.214	-.180	-.113	-.045	.028					
.550	-.160	-.143	-.105	-.056	-.017					
.650	-.093	-.092	-.068	-.036	-.031	-.090	-.066	-.061	-.074	-.119
.750	-.021	-.023	-.014	-.011	-.049	-.048	-.012	-.039	-.054	-.121
.850	.063	.066	.067	.054	-.037	.013	.055	.032	-.014	-.102
.900	.099	.095	.087	.070	-.068	.023	.069	.038	-.016	-.128

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(h)  $\alpha = 0^\circ$ ;  $M = 0.85$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.743	.682	.589	.076	-.137	.538	.787	.593	-.077	-.387
.025	.279	-.057	-.529	-1.682	-1.311	.328	-.076	-1.084	-2.209	-.876
.075	.143	-.088	-.385	-1.402	-1.104	.148	-.124	-.434	-2.014	-.892
.150	.042	-.136	-.340	-1.120	-.920	.049	-.147	-.356	-1.734	-.882
.250	.020	-.096	-.220	-.703	-.711	-.024	-.160	-.298	-.694	-.808
.350	-.037	-.121	-.194	-.601	-.517	-.093	-.181	-.279	-.660	-.721
.450	-.096	-.144	-.181	-.545	-.408	-.142	-.196	-.252	-.630	-.616
.550	-.089	-.108	-.119	-.473	-.321	-.131	-.161	-.174	-.538	-.516
.650	-.069	-.073	-.066	-.411	-.209	-.105	-.113	-.102	-.437	-.434
.750	-.030	-.013	.002	-.346	-.140	-.043	-.017	-.029	-.333	-.341
.850	.028	.056	.062	-.254	-.057	.043	.060	.056	-.231	-.240
.900	.062	.092	.093	-.211	-.011	.078	.097	.096	-.183	-.177
	<i>Right side</i>									
.025	-.384	.019	.339	.455	.833	-.971	-.037	.366	.435	.750
.075	-.398	-.086	.142	.141	.560	-.431	-.114	.155	.139	.537
.150	-.326	-.111	.060	-.019	.395	-.344	-.131	.054	-.028	.381
.250	-.223	-.098	.020	-.120	.281	-.279	-.147	-.018	-.161	.260
.350	-.201	-.126	-.044	-.224	.174	-.257	-.174	-.080	-.260	.151
.450	-.174	-.143	-.093	-.315	.080	-.135	-.100	-.080	-.373	-.036
.550	-.115	-.106	-.083	-.330	.047	-.165	-.147	-.119	-.378	.004
.650	-.050	-.061	-.058	-.319	.022	-.088	-.088	-.086	-.355	-.016
.750	.003	-.011	-.029	-.308	.009	.000	-.007	-.019	-.296	-.009
.850	.071	.066	.044	-.227	.042	.081	.074	.063	-.220	.014
.900	1.187	1.192	1.192	1.182	1.184	.110	.103	.079	-.202	-.002
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.477	.763	.533	-.150	-.306	.531	.720	.442	-.245	-.243
.025	.340	-.110	-1.180	-2.279	-.689	.200	-.256	-1.087	-1.328	-.477
.075	.156	-.153	-.560	-2.093	-.698	-.014	-.204	-.686	-1.283	-.490
.150	.035	-.178	-.411	-1.493	-.702	-.098	-.177	-.297	-1.265	-.557
.250	-.050	-.190	-.316	-1.209	-.658	-.139	-.191	-.253	-1.084	-.543
.350	-.113	-.204	-.268	-.928	-.629					
.450	-.159	-.207	-.240	-.674	-.577					
.550	-.151	-.170	-.187	-.534	-.517					
.650	-.103	-.110	-.108	-.425	-.465	-.096	-.066	-.121	-.728	-.385
.750	-.054	-.011	-.029	-.335	-.408	-.036	-.003	-.069	-.671	-.359
.850	.041	.049	.041	-.258	-.349	.010	.042	.037	-.610	-.328
.900	.055	.069	.070	-.227	-.319	.026	.069	-.021	-.570	-.315
	<i>Right side</i>									
.025	-1.149	-.050	.382	.428	.712	-1.108	-.150	.242	.223	.530
.075	-.436	-.137	.153	.133	.512	-.470	-.204	-.032	-.154	.263
.150	-.362	-.143	.051	-.037	.359	-.264	-.178	-.102	-.349	.088
.250	-.303	-.180	-.044	-.190	.220					
.350	-.266	-.196	-.113	-.314	.103					
.450	-.227	-.191	-.148	-.391	.018					
.550	-.168	-.158	-.142	-.411	-.036					
.650	-.093	-.096	-.090	-.382	-.051	-.089	-.070	-.077	-.427	-.141
.750	-.016	-.013	-.031	-.330	-.062	-.044	-.007	-.050	-.389	-.135
.850	.078	.079	.062	-.229	-.042	.019	.072	.021	-.321	-.111
.900	.111	.107	.083	-.208	-.069	.030	.083	.036	-.321	-.134

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND  
VERTICAL TAIL - Continued

(i)  $\alpha = 0^\circ$ ;  $M = 0.90$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.775	.693	.645	.419	.025	.600	.792	.664	.335	-.189
.025	.284	-.050	-.462	-.939	-1.114	.318	-.084	-.944	-1.301	-.743
.075	.147	-.089	-.383	-.769	-.971	.139	-.140	-.610	-1.169	-.755
.150	.030	-.156	-.378	-.645	-.828	.039	-.162	-.395	-1.053	-.769
.250	.011	-.118	-.241	-.494	-.659	-.041	-.185	-.324	-.610	-.749
.350	-.052	-.146	-.218	-.331	-.564	-.117	-.226	-.331	-.453	-.689
.450	-.114	-.180	-.206	-.260	-.508	-.194	-.256	-.269	-.296	-.614
.550	-.121	-.149	-.135	-.143	-.425	-.177	-.202	-.174	-.106	-.534
.650	-.101	-.097	-.068	-.067	-.330	-.126	-.134	-.101	-.043	-.468
.750	-.048	-.023	.010	-.002	-.229	-.053	-.029	-.019	.018	-.394
.850	.021	.050	.069	.057	-.133	.031	.055	.065	.087	-.314
.900	.057	.087	.101	.088	-.070	.074	.099	.109	.124	-.254
	<i>Right side</i>									
.025	-.337	.025	.336	.600	.833	-.890	-.044	.352	.581	.745
.075	-.411	-.093	.148	.361	.567	-.616	-.129	.150	.353	.532
.150	-.379	-.140	.044	.218	.396	-.403	-.154	.048	.217	.376
.250	-.260	-.122	.009	.141	.279	-.323	-.174	-.030	.108	.253
.350	-.237	-.152	-.056	.055	.166	-.324	-.213	-.100	.018	.138
.450	-.216	-.181	-.116	-.027	.063	-.200	-.177	-.131	-.065	.009
.550	-.132	-.145	-.112	-.049	.019	-.178	-.193	-.164	-.097	-.031
.650	-.061	-.085	-.080	-.040	-.004	-.094	-.117	-.109	-.073	-.056
.750	.006	-.018	-.029	-.008	-.020	-.005	-.010	-.023	-.008	-.045
.850	.078	.065	.048	.061	.008	.085	.078	.061	.067	-.025
.900	.101	.098	.069	.076	.009	.115	.106	.085	.086	-.046
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.532	.771	.602	.271	-.158	.564	.717	.493	.141	-.201
.025	.317	-.125	-1.058	-1.377	-.719	.171	-.299	-1.021	-1.325	-.487
.075	.137	-.174	-.874	-1.243	-.731	-.064	-.284	-1.013	-1.329	-.497
.150	.014	-.210	-.406	-1.150	-.734	-.197	-.226	-.437	-1.079	-.557
.250	-.081	-.232	-.407	-1.090	-.636	-.188	-.232	-.178	-.741	-.557
.350	-.166	-.260	-.264	-.682	-.606					
.450	-.217	-.258	-.226	-.124	-.586					
.550	-.198	-.214	-.191	-.064	-.556					
.650	-.129	-.121	-.105	-.024	-.515	-.109	-.076	-.112	-.261	-.411
.750	-.050	-.025	-.021	.018	-.463	-.038	-.009	-.050	-.255	-.390
.850	.026	.043	.030	.063	-.399	-.001	.041	-.027	-.233	-.362
.900	.046	.073	.084	.084	-.372	.018	.066	-.015	-.203	-.346
	<i>Right side</i>									
.025	-1.017	-.058	.360	.572	.709	-.997	-.174	.219	.410	.528
.075	-.810	-.156	.140	.337	.505	-.955	-.293	-.077	.098	.261
.150	-.387	-.160	.040	.205	.355	-.377	-.230	-.190	-.088	.069
.250	-.404	-.221	-.068	.074	.213					
.350	-.313	-.258	-.152	-.040	.080					
.450	-.225	-.242	-.197	-.116	-.016					
.550	-.185	-.194	-.172	-.131	-.076					
.650	-.100	-.109	-.108	-.090	-.094	.055	.001	-.022	-.049	-.097
.750	-.019	-.025	-.040	-.028	-.102	-.050	-.010	-.049	-.051	-.166
.850	.079	.077	.056	.062	-.070	.018	.062	.029	.008	-.134
.900	.110	.113	.089	.090	-.097	.031	.083	.042	.018	-.152

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(j)  $\alpha = 0^\circ$ ;  $M = 0.92$

$x$ $c_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.781	.698	.658	.446	.072	.613	.802	.681	.366	-.142
.025	.286	-.036	-.440		-1.092	.317	-.067	-.912	-1.253	-.726
.075	.152	-.079	-.375	-.754	-.950	.140	-.136	-.687	-1.136	-.731
.150	.034	-.149	-.367	-.645	-.815	.040	-.159	-.414	-1.036	-.742
.250	.009	-.114	-.257	-.549	-.648	-.039	-.181	-.324	-.674	-.730
.350	-.053	-.146	-.224	-.373	-.558	-.119	-.229	-.345	-.520	-.684
.450	-.129	-.185	-.223	-.311	-.514	-.201	-.268	-.332	-.420	-.619
.550	-.134	-.156	-.141	-.175	-.441	-.197	-.213	-.159	-.120	-.545
.650	-.106	-.095	-.064	-.077	-.354	-.142	-.140	-.082	-.032	-.477
.750	-.053	-.020	.012	-.005	-.249	-.051	-.020	-.006	.030	-.405
.850	.023	.041	.080	.059	-.151	.035	.045	.075	.097	-.325
.900	.060	.097	.109	.093	-.095	.082	.110	.117	.125	-.273
<i>Right side</i>										
.025	-.324	.036	.343	.612	.835	-.876	-.035	.351	.582	.747
.075	-.410	-.084	.149	.364	.566	-.645	-.123	.149	.353	.532
.150	-.379	-.132	.054	.224	.396	-.427	-.149	.054	.214	.375
.250	-.271	-.119	.011	.147	.282	-.321	-.172	-.026	.104	.249
.350	-.241	-.149	-.054	.054	.168	-.338	-.211	-.098	.014	.139
.450	-.226	-.183	-.122	-.037	.059	-.250	-.215	-.153	-.077	.017
.550	-.136	-.149	-.124	-.058	.017	-.163	-.200	-.183	-.116	-.036
.650	-.055	-.078	-.087	-.051	-.019	-.080	-.109	-.119	-.096	-.068
.750	.014	-.012	-.034	-.026	-.037	.008	-.007	-.024	-.019	-.058
.850	.078	.074	.049	.053	-.004	.092	.087	.068	.064	-.037
.900	.109	.102	.074	.074	-.015	.120	.114	.095	.083	-.062
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.542	.772	.620	.302	-.117	.571	.724	.518	.183	-.176
.025	.321	-.110	-1.014	-1.325	-.737	.170	-.275	-.967	-1.282	-.486
.075	.140	-.165	-.864	-1.200	-.751	-.063	-.298	-.984	-1.282	-.509
.150	.018	-.205	-.602	-1.112	-.728	-.218	-.247	-.738	-1.064	-.562
.250	-.081	-.233	-.390	-1.071	-.699	-.244	-.242	-.185	-.917	-.566
.350	-.177	-.295	-.413	-.962	-.611					
.450	-.247	-.273	-.172	-.358	-.583					
.550	-.209	-.219	-.167	-.043	-.558					
.650	-.131	-.106	-.094	.006	-.526	-.092	-.065	-.111	-.311	-.427
.750	-.042	-.016	-.020	.044	-.473	-.034	-.001	-.051	-.274	-.400
.850	.033	.058	.061	.078	-.413	.006	.053	-.023	-.244	-.370
.900	.055	.087	.091	.098	-.391	.026	.078	-.011	-.219	-.361
<i>Right side</i>										
.025	-.995	-.053	.361	.568	.710	-.983	.167	.220	.411	.536
.075	-.826	-.146	.139	.339	.503	-.960	.316	-.079	.101	.261
.150	-.510	-.158	.041	.202	.352	-.675	.251	-.215	-.095	.067
.250	-.397	-.222	-.071	.067	.208					
.350	-.421	-.293	-.170	-.048	.080					
.450	-.181	-.251	-.231	-.138	-.025					
.550	-.164	-.189	-.185	-.170	-.092					
.650	-.087	-.096	-.111	-.113	-.112	-.083	.058	-.049	-.086	-.150
.750	-.011	-.014	-.032	-.037	-.121	-.055	.004	-.035	-.052	-.181
.850	.086	.089	.070	.063	-.075	.018	.076	.038	.014	-.146
.900	.120	.123	.104	.091	-.106	.030	.093	.057	.017	-.164

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued

(k)  $\alpha = 9.4^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.404	.804	.439	-.646	-1.194	.162	.693	.174	-.716	-1.505
.025	-.031	-.132	-.231	-.734	-2.578	.333	-.109	-.701	-1.213	-1.315
.075	.015	-.136	-.316	-.535	-.893	.149	-.141	-.463	-1.045	-1.265
.150	-.019	-.164	-.314	-.444	-.664	.054	-.143	-.341	-.793	-1.333
.250	-.019	-.109	-.225	-.319	-.427	-.012	-.155	-.282	-.439	-1.176
.350	-.044	-.116	-.206	-.265	-.359	-.065	-.166	-.259	-.317	-.806
.450	-.065	-.132	-.195	-.231	-.345	-.095	-.180	-.234	-.274	-.457
.550	-.067	-.116	-.158	-.190	-.258	-.093	-.141	-.179	-.213	-.308
.650	-.044	-.079	-.117	-.149	-.180	-.070	-.104	-.126	-.154	-.240
.750	-.015	-.038	-.062	-.074	-.114	-.035	-.063	-.062	-.086	-.164
.850	.015	.011	.002	-.013	-.037	.020	.008	.013	-.009	-.078
.900	.041	.045	.039	.016	.007	.045	.045	.045	.025	-.034
<i>Right side</i>										
.025	-.178	-.044	.000	.504	.806	-.684	-.063	.373	.583	.715
.075	-.327	-.136	.013	.320	.543	-.451	-.116	.162	.363	.536
.150	-.309	-.146	-.016	.214	.383	-.343	-.129	.064	.227	.383
.250	-.238	-.102	-.028	.150	.281	-.281	-.141	.004	.134	.267
.350	-.221	-.118	-.048	.073	.180	-.256	-.157	-.046	.062	.178
.450	-.203	-.132	-.074	.019	.107	-.231	-.157	-.078	.003	.100
.550	-.159	-.107	-.060	.005	.073	-.194	-.136	-.080	-.020	.059
.650	-.109	-.072	-.037	.010	.062	-.125	-.093	-.051	-.013	.039
.750	-.056	-.024	-.005	.019	.057	-.054	-.031	-.014	.016	.048
.850	.011	.041	.039	.062	.087	.025	.038	.050	.057	.078
.900	.045	.061	.064	.071	.082	.054	.050	.055	.053	.055
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.041	.643	.034	-.841	-1.110	.188	.585	-.039	-.485	-.811
.025	.337	-.139	-.977	-1.369	-.897	.199	-.226	-.872	-.730	-.594
.075	.156	-.159	-.517	-1.340	-.884	.006	-.187	-.453	-.711	-.587
.150	.043	-.164	-.387	-1.176	-.849	-.088	-.173	-.286	-.673	-.594
.250	-.026	-.164	-.300	-.639	-.813	-.079	-.104	-.197	-.573	-.578
.350	-.074	-.173	-.261	-.315	-.781					
.450	-.113	-.178	-.227	-.242	-.738					
.550	-.125	-.164	-.190	-.210	-.664					
.650	-.090	-.109	-.131	-.151	-.575	-.111	-.072	-.122	-.340	-.461
.750	-.044	-.051	-.064	-.086	-.480	-.072	-.024	-.087	-.297	-.434
.850	-.003	-.003	-.007	-.036	-.379	-.035	.013	-.058	-.253	-.416
.900	.015	.025	.018	-.011	-.336	-.019	.027	-.037	-.231	-.395
<i>Right side</i>										
.025	-.881	-.065	.380	.570	.644	-.718	-.127	.249	.375	.450
.075	-.502	-.129	.162	.361	.495	-.417	-.178	.007	.112	.237
.150	-.350	-.132	.064	.220	.356	-.270	-.152	-.074	-.015	.087
.250	-.300	-.157	-.016	.103	.228					
.350	-.272	-.173	-.071	.023	.135					
.450	-.231	-.166	-.099	-.020	.062					
.550	-.189	-.141	-.096	-.045	.020					
.650	-.125	-.107	-.071	-.036	.000	-.097	-.063	-.085	-.115	-.130
.750	-.061	-.031	-.030	-.024	-.012	-.074	-.024	-.071	-.106	-.151
.850	.015	.031	.032	.030	.000	-.026	.025	-.009	-.070	-.149
.900	.045	.054	.055	.044	-.021	-.003	.048	.004	-.056	-.171

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND  
VERTICAL TAIL - Continued

(t)  $\alpha = 9.6^\circ$ ;  $M = 0.80$

$x$ $C_V$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_V = 0.11$					$z/b_V = 0.38$				
	<i>Left side</i>									
.000	.356	.848	.422	-.349	-.188	.404	.723	.434	-.137	-.801
.025	-.043	-.119	-.153	-.597	-1.595	.328	-.106	-1.046	-1.517	-1.046
.075	.020	-.134	-.325	-.569	-1.058	.151	-.149	-.499	-1.094	-1.028
.150	-.016	-.176	-.363	-.537	-.705	.057	-.148	-.381	-.770	-.985
.250	-.022	-.121	-.256	-.374	-.485	-.015	-.163	-.317	-.490	-.930
.350	-.056	-.130	-.232	-.307	-.404	-.081	-.198	-.299	-.393	-.797
.450	-.081	-.154	-.224	-.272	-.360	-.127	-.213	-.270	-.327	-.631
.550	-.081	-.134	-.182	-.219	-.307	-.115	-.167	-.201	-.244	-.497
.650	-.059	-.095	-.130	-.169	-.233	-.091	-.121	-.140	-.169	-.397
.750	-.024	-.045	-.063	-.090	-.163	-.043	-.056	-.063	-.082	-.297
.850	.019	.020	.006	-.009	-.084	.025	.024	.023	-.001	-.183
.900	.051	.057	.045	.027	-.038	.060	.066	.064	.043	-.119
	<i>Right side</i>									
.025	-.095	-.018	-.025	.390	.768	-1.077	-.048	.375	.589	.715
.075	-.333	-.131	.024	.312	.521	-.468	-.116	.169	.370	.532
.150	-.352	-.149	.003	.218	.375	-.367	-.136	.064	.237	.393
.250	-.262	-.110	-.020	.154	.274	-.300	-.148	.003	.137	.271
.350	-.241	-.130	-.052	.079	.172	-.276	-.169	-.055	.059	.180
.450	-.229	-.151	-.083	.005	.091	-.250	-.184	-.101	-.009	.088
.550	-.175	-.125	-.072	-.015	.056	-.198	-.161	-.112	-.044	.038
.650	-.116	-.080	-.044	-.008	.044	-.125	-.109	-.073	-.029	.021
.750	-.059	-.035	-.012	.002	.036	-.040	-.035	-.022	.002	.026
.850	.023	.041	.045	.050	.064	.045	.047	.045	.053	.045
.900	.058	.068	.067	.064	.058	.073	.066	.061	.059	.021
	$z/b_V = 0.66$					$z/b_V = 0.93$				
	<i>Left side</i>									
.000	.284	.668	.295	-.272	-.711	.321	.600	.178	-.256	-.622
.025	.336	-.140	-1.197	-1.250	-.711	.204	-.259	-.895	-.685	-.541
.075	.155	-.164	-.650	-1.173	-.714	-.015	-.232	-.680	-.664	-.523
.150	.043	-.182	-.439	-1.043	-.700	-.122	-.191	-.354	-.671	-.523
.250	-.036	-.188	-.325	-.871	-.671	-.086	-.115	-.237	-.600	-.535
.350	-.095	-.204	-.285	-.619	-.650					
.450	-.136	-.207	-.250	-.387	-.627					
.550	-.144	-.179	-.203	-.248	-.590					
.650	-.103	-.118	-.131	-.149	-.552	-.106	-.069	-.115	-.358	-.445
.750	-.043	-.044	-.054	-.062	-.500	-.053	-.012	-.072	-.310	-.422
.850	.010	.018	.012	.000	-.445	-.010	.033	-.037	-.265	-.392
.900	.037	.047	.042	.029	-.412	.004	.053	-.017	-.233	-.361
	<i>Right side</i>									
.025	-1.103	-.059	.379	.574	.666	-.943	-.134	.256	.406	.483
.075	-.572	-.134	.166	.364	.503	-.578	-.214	-.022	.134	.263
.150	-.383	-.142	.071	.231	.366	-.277	-.167	-.096	-.015	.105
.250	-.323	-.170	-.022	.111	.238					
.350	-.283	-.195	-.086	.023	.132					
.450	-.241	-.190	-.116	-.037	.052					
.550	-.189	-.164	-.125	-.062	-.003					
.650	-.121	-.109	-.086	-.050	-.026	-.089	-.054	-.081	-.113	-.154
.750	-.045	-.039	-.031	-.023	-.043	-.059	-.015	-.052	-.096	-.169
.850	.043	.048	.045	.037	-.032	-.001	.051	.012	-.049	-.159
.900	.076	.077	.070	.055	-.064	.022	.069	.030	-.050	-.180



TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continued  
(m)  $\alpha = 9.7^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.336	.866	.415	-.203	-.029	.470	.740	.511	.018	-.622
.025	-.065	-.108	-.103	-.519	-1.411	.325	-.100	-1.153	-1.579	-1.016
.075	.016	-.129	-.314	-.539	-.996	.146	-.149	-.476	-1.407	-1.032
.150	-.020	-.177	-.397	-.583	-.736	.057	-.158	-.392	-.666	-.972
.250	-.031	-.117	-.273	-.411	-.508	-.017	-.172	-.335	-.439	-.884
.350	-.066	-.134	-.247	-.318	-.433	-.088	-.209	-.317	-.382	-.767
.450	-.091	-.165	-.247	-.271	-.386	-.142	-.228	-.288	-.350	-.643
.550	-.091	-.148	-.199	-.224	-.338	-.130	-.189	-.214	-.261	-.545
.650	-.068	-.103	-.133	-.167	-.266	-.103	-.128	-.140	-.174	-.458
.750	-.037	-.050	-.070	-.093	-.200	-.054	-.059	-.062	-.085	-.365
.850	.019	.023	.005	-.004	-.117	.020	.029	.022	.009	-.245
.900	.051	.066	.046	.041	-.074	.064	.073	.071	.051	-.177
<i>Right side</i>										
.025	-.049	-.015	-.051	.327	.728	-1.044	-.049	.361	.586	.719
.075	-.326	-.131	.012	.294	.488	-.462	-.122	.155	.370	.532
.150	-.370	-.162	-.008	.220	.347	-.373	-.138	.062	.242	.388
.250	-.276	-.111	-.038	.165	.256	-.315	-.158	-.013	.142	.268
.350	-.255	-.136	-.068	.081	.163	-.292	-.182	-.073	.065	.169
.450	-.244	-.166	-.103	.009	.079	-.265	-.197	-.116	-.012	.076
.550	-.187	-.136	-.088	-.015	.044	-.206	-.179	-.129	-.052	.025
.650	-.120	-.088	-.059	-.009	.029	-.133	-.115	-.093	-.041	.000
.750	-.064	-.042	-.029	.002	.018	-.040	-.036	-.033	-.005	.000
.850	.024	.039	.041	.051	.041	.048	.053	.042	.055	.018
.900	.063	.070	.058	.071	.032	.081	.076	.061	.062	-.012
$z/b_v = 0.66$										
<i>Left side</i>										
.000	.342	.680	.365	-.135	-.619	.351	.609	.207	-.240	-.613
.025	.331	-.132	-1.253	-1.365	-.702	.197	-.258	-1.163	-.764	-.545
.075	.152	-.168	-.659	-1.257	-.703	-.031	-.265	-.721	-.731	-.528
.150	.036	-.189	-.466	-1.016	-.692	-.150	-.207	-.428	-.749	-.532
.250	-.049	-.200	-.355	-.794	-.673	-.102	-.118	-.240	-.666	-.545
.350	-.113	-.220	-.296	-.613	-.650					
.450	-.154	-.223	-.264	-.412	-.631					
.550	-.170	-.194	-.213	-.264	-.598					
.650	-.119	-.121	-.136	-.146	-.564	-.110	-.073	-.125	-.347	-.474
.750	-.048	-.043	-.060	-.055	-.518	-.054	-.011	-.076	-.290	-.441
.850	.007	.022	.014	.009	-.463	-.010	.039	-.039	-.237	-.413
.900	.040	.054	.046	.045	-.439	.010	.062	-.023	-.206	-.400
<i>Right side</i>										
.025	-1.235	-.059	.367	.571	.671	-1.152	-.145	.242	.411	.493
.075	-.533	-.138	.155	.364	.509	-.713	-.252	-.058	.132	.268
.150	-.398	-.149	.062	.236	.372	-.289	-.177	-.130	-.032	.106
.250	-.342	-.185	-.036	.118	.240					
.350	-.302	-.214	-.102	.022	.129					
.450	-.254	-.209	-.147	-.052	.035					
.550	-.198	-.180	-.145	-.079	-.023					
.650	-.122	-.111	-.106	-.062	-.049	-.092	-.054	-.098	-.117	-.182
.750	-.042	-.037	-.049	-.028	-.069	-.059	-.012	-.066	-.098	-.195
.850	.047	.056	.039	.045	-.054	.010	.059	.011	-.045	-.182
.900	.081	.084	.066	.062	-.089	.027	.079	.025	-.043	-.205

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Continue I

(n)  $\alpha = 9.7^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.300	.911	.414	-.105	.098	.531	.752	.572	.157	-.465
.025	-.092	-.081	-.019	-.446	-1.267	.324	-.088	-1.037	-1.403	-.944
.075	.017	-.109	-.248	-.485	-.927	.150	-.142	-.568	-1.266	-.955
.150	-.013	-.167	-.415	-.583	-.700	.054	-.155	-.473	-.752	-.897
.250	-.026	-.114	-.297	-.517	-.571	-.021	-.178	-.341	-.660	-.826
.350	-.066	-.141	-.265	-.421	-.488	-.092	-.224	-.361	-.504	-.755
.450	-.098	-.179	-.274	-.337	-.437	-.162	-.268	-.345	-.286	-.654
.550	-.110	-.173	-.221	-.296	-.382	-.157	-.215	-.208	-.266	-.566
.650	-.089	-.120	-.142	-.171	-.313	-.126	-.143	-.136	-.208	-.496
.750	-.049	-.056	-.068	-.109	-.238	-.065	-.060	-.054	-.105	-.416
.850	.007	.022	.009	-.022	-.149	.021	.029	.034	-.002	-.311
.900	.051	.063	.051	.027	-.096	.066	.075	.081	.046	-.235
<i>Right side</i>										
.025	.038	.016	-.056	.306	.728	-.941	.039	.377	.588	.725
.075	-.276	-.114	.033	.298	.496	-.568	.114	.171	.368	.536
.150	-.411	-.149	.019	.229	.358	-.420	.138	.070	.240	.394
.250	-.292	-.109	-.009	.165	.268	-.329	.158	-.002	.133	.269
.350	-.283	-.139	-.056	.075	.167	-.341	-.203	-.069	.047	.171
.450	-.277	-.184	-.092	-.008	.076	-.305	-.230	-.133	-.040	.062
.550	-.208	-.158	-.092	-.035	.030	-.214	-.204	-.152	-.079	.006
.650	-.129	-.101	-.072	-.037	.010	-.132	-.143	-.114	-.073	-.019
.750	-.062	-.053	-.037	-.026	-.004	-.038	-.040	-.042	-.029	-.022
.850	.026	.037	.034	.035	.014	.055	.053	.042	.035	-.008
.900	.062	.066	.057	.050	.006	.085	.072	.059	.044	-.030
$z/b_v = 0.66$										
<i>Left side</i>										
.000	.404	.684	.436	.018	-.512	.390	.616	.278	-.129	-.553
.025	.328	-.129	-1.169	-1.424	-.668	.183	-.256	-1.162	-.768	-.528
.075	.150	-.169	-.986	-1.313	-.673	-.053	-.314	-1.074	-.782	-.514
.150	.029	-.204	-.456	-1.163	-.677	-.246	-.233	-.620	-.516	-.516
.250	-.060	-.226	-.415	-.770	-.664	-.134	-.125	-.194	-.758	-.528
.350	-.138	-.265	-.393	-.612	-.644					
.450	-.200	-.263	-.229	-.509	-.620					
.550	-.197	-.218	-.206	-.397	-.589					
.650	-.134	-.128	-.130	-.243	-.559	-.118	-.076	-.122	-.399	-.475
.750	-.057	-.047	-.046	-.113	-.523	-.054	-.008	-.074	-.337	-.453
.850	.010	.023	.022	-.016	-.475	-.006	.043	-.033	-.281	-.425
.900	.041	.058	.055	.026	-.448	.014	.059	-.011	-.251	-.409
<i>Right side</i>										
.025	-1.120	-.051	.370	.562	.674	-1.150	-.141	.246	.403	.502
.075	-.901	-.141	.163	.353	.509	-1.036	-.318	-.061	.119	.280
.150	-.364	-.151	.066	.226	.374	-.372	-.206	-.202	-.065	.108
.250	-.409	-.208	-.041	.099	.241					
.350	-.356	-.243	-.125	-.005	.123					
.450	-.238	-.253	-.186	-.091	.026					
.550	-.201	-.207	-.174	-.131	-.042					
.650	-.121	-.129	-.122	-.109	-.073	-.098	-.061	-.100	-.159	-.199
.750	-.044	-.041	-.053	-.060	-.087	-.061	-.016	-.064	-.125	-.213
.850	.054	.057	.041	.019	-.073	.006	.061	.015	-.068	-.193
.900	.090	.084	.070	.040	-.102	.026	.078	.030	-.064	-.208

TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(a)  $\alpha = 9.8^\circ$ ;  $M = 0.92$ 

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.313	.968	.449	-.173		.560	.749	.591	.190	
.025	-.059	-.034	.048	-.363		.329	-.038	-.978	-1.331	
.075	.044	-.063	-.164	-.408		.161	-.097	-.420	-1.189	
.150	.027	-.119	-.328	-.519		.062	-.119	-.375	-.687	
.250	.002	-.085	-.240	-.484		-.015	-.153	-.323	-.630	
.350	-.044	-.119	-.237	-.423		-.094	-.200	-.349	-.547	
.450	-.088	-.170	-.265	-.362		-.161	-.251	-.365	-.306	
.550	-.103	-.175	-.222	-.253		-.169	-.217	-.211	-.270	
.650	-.090	-.127	-.138	-.176		-.133	-.152	-.135	-.218	
.750	-.049	-.063	-.070	-.106		-.069	-.067	-.057	-.112	
.850	.010	.014	.008	-.024		.012	.023	.030	-.007	
.900	.045	.056	.052	.020		.058	.074	.076	.040	
	<i>Right side</i>									
.025	.121	.058	.004	.233		-.855	.000	.397	.593	
.075	-.185	-.068	.071	.300		-.458	-.081	.192	.379	
.150	-.333	-.110	.063	.262		-.370	-.108	.096	.249	
.250	-.253	-.084	.033	.183		-.310	-.135	.017	.137	
.350	-.260	-.124	-.016	.083		-.340	-.188	-.057	.047	
.450	-.274	-.172	-.070	-.005		-.333	-.225	-.122	-.042	
.550	-.212	-.163	-.084	-.040		-.214	-.206	-.148	-.093	
.650	-.132	-.108	-.067	-.048		-.135	-.142	-.116	-.093	
.750	-.072	-.062	-.037	-.039		-.042	-.045	-.043	-.047	
.850	.023	.031	.030	.022		.048	.047	.039	.022	
.900	.057	.065	.056	.040		.081	.069	.056	.032	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.414	.686	.438	.049		.413	.615	.299	-.085	
.025	.325	-.084	-1.116	-1.409		.182	-.235	-1.136	-.777	
.075	.152	-.131	-.920	-1.285		-.051	-.288	-1.063	-.772	
.150	.036	-.168	-.408	-1.132		-.245	-.222	-.506	-.789	
.250	-.057	-.204	-.416	-.764		-.144	-.115	-.177	-.755	
.350	-.143	-.258	-.425	-.620						
.450	-.208	-.269	-.223	-.517						
.550	-.204	-.217	-.197	-.406						
.650	-.139	-.133	-.131	-.258		-.123	-.078	-.125	-.402	
.750	-.066	-.049	-.051	-.124		-.056	-.007	-.072	-.338	
.850	.006	.023	.019	-.019		-.005	.041	-.030	-.279	
.900	.033	.056	.052	.024		.012	.062	-.009	-.250	
	<i>Right side</i>									
.025	-1.052	-.030	.383	.558		-1.093	-.129	.251	.405	
.075	-.781	-.114	.181	.355		-.972	-.300	-.051	.117	
.150	-.365	-.131	.080	.227		-.349	-.189	-.189	-.063	
.250	-.399	-.191	-.029	.096						
.350	-.413	-.240	-.119	-.010						
.450	-.231	-.248	-.186	-.099						
.550	-.202	-.208	-.181	-.151						
.650	-.124	-.148	-.118	-.127		-.095	-.058	-.100	-.158	
.750	-.042	-.030	-.054	-.072		-.056	-.011	-.062	-.130	
.850	.050	.060	.040	.012		.010	.060	.016	-.069	
.900	.083	.086	.069	.035		.029	.078	.034	-.064	



TABLE II. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND

VERTICAL TAIL - Continued

(q)  $\alpha = 15.8^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	-.029	.063	.410	.172	-.497	.311	.776	.360	-.237	-.611
.025	-.993	.072	.343	.061	-.463	-.044	-.069	-.342	-.592	-.737
.075	-.642	-.120	.004	-.123	-.428	-.116	-.165	-.345	-.493	-.670
.150	-.314	-.214	-.205	-.254	-.415	-.104	-.177	-.317	-.430	-.588
.250	-.173	-.159	-.192	-.220	-.360	-.094	-.182	-.301	-.390	-.524
.350	-.102	-.131	-.203	-.213	-.308	-.097	-.205	-.281	-.348	-.460
.450	-.069	-.143	-.211	-.208	-.261	-.091	-.217	-.259	-.302	-.392
.550	-.048	-.135	-.183	-.173	-.214	-.069	-.180	-.211	-.243	-.325
.650	-.029	-.078	-.119	-.116	-.161	-.053	-.140	-.159	-.186	-.269
.750	-.016	-.050	-.084	-.081	-.111	-.037	-.073	-.093	-.120	-.194
.850	-.010	-.005	-.029	-.025	-.054	-.006	-.002	-.028	-.058	-.132
.900	.004	.034	-.005	.007	-.018	-.186	-.161	-.134	-.142	-.136
<i>Right side</i>										
.025	.354	-.754	-.971	-.483	-.307	-.465	-.288	-.054	.016	.040
.075	-.051	-.435	-.653	-.260	-.185	-.404	-.277	-.110	-.025	.082
.150	-.246	-.262	-.314	-.169	-.074	-.363	-.221	-.090	-.020	.140
.250	-.237	-.156	-.156	-.090	.032	-.328	-.170	-.063	.011	.190
.350	-.252	-.099	-.075	-.031	.096	-.301	-.155	-.044	.027	.204
.450	-.241	-.084	-.032	.008	.131	-.281	-.159	-.047	.037	.201
.550	-.206	-.062	-.011	.034	.129	-.235	-.131	-.029	.045	.183
.650	-.145	-.029	.006	.051	.116	-.176	-.084	-.011	.055	.163
.750	-.108	-.010	.020	.054	.102	-.110	-.031	.017	.061	.134
.850	-.042	.034	.045	.063	.085	-.044	.036	.045	.078	.114
.900	-.026	.049	.041	.063	.075	-.022	.048	.039	.066	.088
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.276	.671	.425	-.011	-.478	.202	.577	.117	-.351	-.665
.025	.270	-.112	-1.093	-1.272	-.822	.149	-.241	-1.177	-.783	-.556
.075	.105	-.161	-.578	-1.142	-.802	-.035	-.224	-.616	-.752	-.538
.150	.016	-.182	-.432	-.864	-.776	-.127	-.174	-.302	-.681	-.527
.250	-.034	-.187	-.330	-.530	-.719	-.076	-.088	-.224	-.593	-.503
.350	-.072	-.202	-.282	-.357	-.633					
.450	-.102	-.206	-.256	-.272	-.530					
.550	-.107	-.185	-.215	-.208	-.433					
.650	-.076	-.128	-.147	-.139	-.348	-.099	-.053	-.116	-.317	-.371
.750	-.041	-.059	-.072	-.073	-.275	-.072	-.013	-.084	-.255	-.343
.850	-.009	-.002	-.008	-.026	-.214	-.047	.033	-.052	-.198	-.308
.900	.006	.030	.023	.001	-.179	-.025	.048	-.020	-.163	-.287
<i>Right side</i>										
.025	-1.186	-.140	.335	.540	.628	-.934	-.170	.215	.372	.474
.075	-.598	-.180	.131	.317	.477	-.639	-.218	-.028	.127	.280
.150	-.421	-.147	.056	.214	.383	-.358	-.170	-.113	-.033	.113
.250	-.357	-.158	.000	.133	.312					
.350	-.319	-.173	-.044	.070	.248					
.450	-.272	-.162	-.063	.045	.204					
.550	-.228	-.138	-.061	.037	.161					
.650	-.164	-.088	-.032	.046	.134	-.095	-.043	-.058	-.052	-.038
.750	-.095	-.037	.003	.054	.107	-.118	-.010	-.049	-.054	-.076
.850	-.012	.046	.053	.084	.084	-.054	.056	.015	-.004	-.074
.900		.069	.068	.077	.057		.068	.024	-.004	-.091

TABLE II.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, AND VERTICAL TAIL - Concluded

(r)  $\alpha = 15.9^\circ$ ;  $M = 0.85$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.392	.914	.581	.204	-.613	.429	.776	.400	-.250	-.585
.025	-1.096	-.065	.261	-.067	-.582	.200	-.174	-.483	-.666	-.740
.075	-.538	-.258	-.045	-.225	-.525	.013	-.252	-.454	-.570	-.682
.150	-.260	-.322	-.291	-.354	-.497	-.059	-.240	-.403	-.510	-.611
.250	-.185	-.225	-.266	-.297	-.434	-.095	-.217	-.356	-.465	-.551
.350	-.149	-.162	-.245	-.274	-.366	-.132	-.232	-.330	-.409	-.484
.450	-.121	-.162	-.242	-.253	-.301	-.145	-.241	-.291	-.348	-.417
.550	-.091	-.142	-.205	-.200	-.235	-.114	-.190	-.232	-.283	-.352
.650	-.048	-.076	-.124	-.136	-.174	-.085	-.143	-.175	-.215	-.292
.750	-.020	-.045	-.092	-.094	-.123	-.046	-.078	-.104	-.138	-.218
.850	.013	.000	-.031	-.031	-.062	.008	-.007	-.032	-.067	-.154
.900	.035	.033	.000	-.003	-.025	-.162	-.162	-.138	-.141	-.136
	<i>Right side</i>									
.025	.269	-.620	-1.099	-.543	-.433	-1.102	-.285	-.052	-.047	-.035
.075	-.172	-.803	-.718	-.338	-.302	-.770	-.323	-.147	-.101	-.005
.150	-.466	-.335	-.354	-.253	-.175	-.550	-.276	-.153	-.095	.051
.250	-.333	-.241	-.229	-.175	-.057	-.326	-.224	-.127	-.061	.116
.350	-.260	-.164	-.162	-.105	.022	-.297	-.203	-.107	-.040	.139
.450	-.228	-.129	-.104	-.053	.075	-.265	-.194	-.099	-.021	.153
.550	-.177	-.091	-.059	-.014	.096	-.215	-.156	-.073	-.002	.150
.650	-.115	-.041	-.021	.011	.092	-.148	-.102	-.041	.021	.139
.750	-.063	-.014	.006	.030	.088	-.071	-.040	-.005	.040	.119
.850	.005	.031	.032	.050	.076	.004	.030	.036	.062	.099
.900	.025	.047	.040	.048	.064	.034	.041	.033	.048	.072
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.302	.665	.451	.102	-.316	.256	.576	.175	-.293	-.639
.025	.315	-.153	-1.209	-1.375	-.811	.160	-.268	-1.147	-.869	-.548
.075	.134	-.194	-.670	-1.192	-.794	-.039	-.268	-.696	-.827	-.544
.150	.020	-.211	-.478	-.875	-.770	-.162	-.198	-.369	-.737	-.537
.250	-.053	-.213	-.358	-.611	-.724	-.082	-.113	-.228	-.644	-.518
.350	-.105	-.224	-.300	-.439	-.649					
.450	-.142	-.224	-.272	-.315	-.551					
.550	-.141	-.198	-.226	-.236	-.460					
.650	-.103	-.139	-.151	-.158	-.374	-.095	-.057	-.120	-.331	-.386
.750	-.045	-.064	-.073	-.087	-.301	-.058	-.015	-.080	-.271	-.356
.850	.003	.002	-.002	-.029	-.237	-.019	.027	-.041	-.210	-.320
.900	.025	.030	.027	-.002	-.211	.003	.053	-.019	-.175	-.303
	<i>Right side</i>									
.025	-1.369	-.162	.292	.513	.643	-1.227	-.200	.187	.361	.486
.075	-1.059	-.214	.088	.284	.466	-.742	-.264	-.073	.105	.283
.150	-.396	-.187	.015	.182	.372	-.409	-.196	-.161	-.073	.106
.250	-.323	-.197	-.046	.098	.287					
.350	-.294	-.208	-.090	.040	.227					
.450	-.252	-.190	-.104	.010	.185					
.550	-.205	-.159	-.094	.001	.149					
.650	-.136	-.101	-.056	.020	.126	-.085	-.049	-.066	-.070	-.050
.750	-.065	-.041	-.018	.037	.099	-.083	-.021	-.056	-.073	-.089
.850	.025	.046	.050	.075	.076	-.018	.058	.016	-.013	-.079
.900	.060	.070	.064	.069	.044	.005	.070	.029	-.017	-.107

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$

(a)  $\alpha = -9.4^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.728	.819	.554	-.150	-1.033	.452	.805	.540	-.109	-.723
.025	.246	-.106	-.558	-.982	-2.182	.338	-.091	-.606	-1.171	-1.074
.075	.105	-.122	-.364	-.606	-1.102	.128	-.146	-.451	-.916	-1.056
.150	-.002	-.168	-.330	-.453	-.684	.025	-.168	-.362	-.531	-.962
.250	-.039	-.137	-.237	-.294	-.403	-.057	-.197	-.326	-.392	-.835
.350	-.066	-.146	-.216	-.239	-.316	-.121	-.226	-.335	-.344	-.679
.450	-.114	-.170	-.216	-.216	-.272	-.162	-.244	-.323	-.314	-.536
.550	-.112	-.150	-.180	-.173	-.221	-.160	-.219	-.264	-.260	-.435
.650	-.085	-.110	-.132	-.114	-.182	-.135	-.177	-.203	-.207	-.369
.750	-.057	-.064	-.066	-.071	-.157	-.094	-.113	-.116	-.146	-.318
.850	.000	-.006	-.009	-.009	-.102	-.009	-.031	-.030	-.071	-.249
.900	.027	.029	.041	.025	-.061	.025	.011	.023	-.023	-.201
	<i>Right side</i>									
.025	-.484	-.064	.283	.545	.781	-.797	-.106	.349	.595	.762
.075	-.418	-.155	.080	.280	.491	-.475	-.168	.125	.328	.505
.150	-.345	-.170	.000	.148	.314	-.379	-.184	.023	.171	.328
.250	-.260	-.146	-.018	.075	.195	-.340	-.204	-.059	.057	.183
.350	-.240	-.161	-.080	-.004	.096	-.336	-.230	-.114	-.025	.089
.450	-.228	-.181	-.118	-.061	.015	-.327	-.244	-.153	-.093	-.005
.550	-.185	-.155	-.109	-.073	-.017	-.279	-.221	-.153	-.112	-.054
.650	-.133	-.106	-.082	-.061	-.024	-.203	-.179	-.130	-.102	-.067
.750	-.073	-.062	-.045	-.041	-.021	-.117	-.099	-.080	-.066	-.056
.850	.011	.005	.018	.016	.011	-.009	-.011	.025	.005	-.024
.900	.039	.038	.046	.037	.015	.018	.007	.032	.005	-.049
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.397	.812	.499	-.116	-.352	.498	.734	.194	-.153	-.375
.025	.363	-.150	-.872	-1.403	-.792	.308	-.496	-1.879	-1.360	-.447
.075	.146	-.204	-.576	-1.066	-.824	-.005	-.543	-1.203	-.451	-.454
.150	.004	-.257	-.512	-.619	-.817	-.228	-.567	-1.150	-.440	-.449
.250	-.105	-.299	-.474	-.440	-.674	-.320	-.534	-.790	-.437	-.463
.350	-.185	-.328	-.471	-.380	-.610					
.450	-.224	-.352	-.440	-.360	-.536					
.550	-.238	-.317	-.371	-.344	-.511					
.650	-.192	-.232	-.271	-.317	-.511	-.235	-.279	-.262	-.444	-.490
.750	-.130	-.153	-.164	-.305	-.502	-.126	-.150	-.157	-.428	-.461
.850	-.041	-.068	-.068	-.276	-.456	-.030	-.040	-.066	-.394	-.412
.900	-.018	-.033	-.023	-.250	-.431	.014	.007	-.020	-.362	-.392
	<i>Right side</i>									
.025	-.831	-.146	.387	.633	.774	-1.416	-.439	.330	.574	.783
.075	-.619	-.226	.121	.344	.510	-1.381	-.545	-.027	.201	.431
.150	-.498	-.250	.000	.173	.330	-1.276	-.547	-.223	-.041	.158
.250	-.491	-.308	-.102	.027	.174					
.350	-.502	-.348	-.180	-.084	.050					
.450	-.448	-.354	-.228	-.157	-.054					
.550	-.384	-.323	-.228	-.189	-.111					
.650	-.281	-.235	-.182	-.178	-.129	-.231	-.230	-.189	-.207	-.196
.750	-.164	-.153	-.098	-.134	-.127	-.139	-.144	-.112	-.191	-.194
.850	-.046	-.040	-.004	-.071	-.111	-.018	-.028	-.002	-.130	-.159
.900	.000	-.006	.023	-.068	-.136	.027	.023	.023	-.143	-.182

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(b)  $\alpha = -9.6^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.849	.883	.720	.367	-.102	.663	.869	.732	.375	-.098
.025	.232	-.098	-.549	-.894	-1.422	.319	-.074	-.988	-1.622	-.976
.075	.095	-.128	-.387	-.797	-1.118	.122	-.151	-.443	-1.406	-.965
.150	-.015	-.187	-.356	-.503	-.881	.012	-.178	-.364	-.384	-.833
.250	-.052	-.158	-.248	-.281	-.601	-.080	-.217	-.330	-.299	-.765
.350	-.095	-.167	-.231	-.229	-.389	-.172	-.273	-.347	-.310	-.595
.450	-.162	-.215	-.240	-.215	-.290	-.248	-.341	-.350	-.296	-.439
.550	-.174	-.197	-.202	-.177	-.229	-.269	-.301	-.289	-.243	-.357
.650	-.151	-.154	-.145	-.125	-.198	-.230	-.223	-.221	-.201	-.320
.750	-.101	-.083	-.073	-.073	-.169	-.138	-.126	-.132	-.148	-.302
.850	-.026	-.005	-.001	-.012	-.113	-.042	-.018	-.037	-.076	-.267
.900	.015	.037	.043	.021	-.073	.009	.032	.018	-.035	-.232
<i>Right side</i>										
.025	-.422	-.028	.286	.541	.773	-.908	.078	.341	.587	.767
.075	-.417	-.147	.082	.283	.491	-.462	-.157	.117	.320	.512
.150	-.360	-.178	-.011	.142	.314	-.376	-.184	.009	.172	.337
.250	-.266	-.152	-.043	.070	.198	-.328	-.221	-.081	.052	.193
.350	-.246	-.186	-.111	-.020	.085	-.337	-.272	-.167	-.052	.076
.450	-.254	-.220	-.173	-.098	-.014	-.342	-.318	-.244	-.147	-.046
.550	-.200	-.195	-.178	-.127	-.063	-.293	-.301	-.262	-.192	-.115
.650	-.136	-.141	-.146	-.118	-.081	-.212	-.227	-.231	-.186	-.137
.750	-.071	-.078	-.099	-.092	-.084	-.118	-.098	-.134	-.127	-.124
.850	.013	.014	-.008	-.020	-.044	-.018	.006	-.015	-.049	-.092
.900	.047	.040	.024	.003	-.032	.021	.029	.000	-.038	-.113
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.666	.896	.743	.392	.007	.755	.840	.677	.311	.061
.025	.346	-.108	-1.117	-1.627	-.845	.225	-.402	-1.181	-1.209	-.403
.075	.134	-.186	-.487	-1.383	-.860	-.079	-.621	-.601	-.349	-.404
.150	-.017	-.258	-.472	-.714	-.793	-.339	-.657	-.548	-.343	-.398
.250	-.150	-.346	-.425	-.467	-.538	-.464	-.728	-.528	-.348	-.400
.350	-.283	-.445	-.437	-.333	-.502					
.450	-.399	-.515	-.432	-.285	-.456					
.550	-.435	-.447	-.371	-.282	-.467					
.650	-.337	-.260	-.291	-.294	-.480	-.339	.255	-.394	-.372	-.427
.750	-.186	-.155	-.218	-.316	-.488	-.213	.117	-.368	-.383	-.416
.850	-.101	-.043	-.140	-.330	-.467	-.130	.006	-.307	-.366	-.384
.900	-.071	.002	-.108	-.325	-.459	-.110	.035	-.271	-.354	-.368
<i>Right side</i>										
.025	-1.097	-.095	.379	.625	.786	-.875	.370	.260	.585	.799
.075	-.481	-.204	.122	.344	.533	-.610	.664	-.100	.201	.453
.150	-.444	-.244	-.006	.178	.353	-.577	.653	-.339	-.073	.164
.250	-.434	-.349	-.155	.013	.183					
.350	-.459	-.445	-.288	-.133	.021					
.450	-.434	-.502	-.388	-.259	-.112					
.550	-.360	-.444	-.426	-.331	-.206					
.650	-.287	-.260	-.345	-.308	-.230	-.360	.232	-.279	-.301	-.272
.750	-.212	-.146	-.178	-.214	-.211	-.354	.104	-.187	-.243	-.247
.850	-.126	-.008	-.058	-.124	-.157	-.295	.009	-.091	-.157	-.169
.900	-.083	.038	-.030	-.128	-.188	-.254	.052	-.076	-.165	-.188



TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(c)  $\alpha = -9.7^\circ$ ;  $M = 0.85$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.880	.907	.758	.468	-.051	.702	.885	.771	.487	-.042
.025	.250	-.105	-.549	-.870	-1.382	.326	-.078	-1.012	-1.417	-.999
.075	.109	-.136	-.426	-.794	-1.083	.130	-.163	-.396	-1.278	-.998
.150	-.016	-.210	-.383	-.656	-.834	.019	-.197	-.376	-1.127	-.825
.250	-.048	-.177	-.263	-.303	-.627	-.078	-.230	-.341	-.227	-.776
.350	-.093	-.192	-.245	-.216	-.424	-.171	-.296	-.355	-.233	-.602
.450	-.174	-.252	-.255	-.200	-.304	-.273	-.404	-.366	-.268	-.439
.550	-.206	-.262	-.212	-.169	-.233	-.320	-.431	-.305	-.233	-.357
.650	-.197	-.223	-.155	-.127	-.202	-.321	-.350	-.232	-.206	-.318
.750	-.143	-.129	-.086	-.080	-.179	-.203	-.161	-.146	-.160	-.306
.850	-.047	-.028	-.004	-.025	-.135	-.061	-.030	-.050	-.093	-.281
.900	.006	.025	.033	.014	-.096	-.006	.029	.009	-.049	-.247
	<i>Right side</i>									
.025	-.414	-.030	.300	.542	.767	-.948	-.081	.348	.581	.760
.075	-.444	-.163	.086	.283	.487	-.527	-.172	.119	.318	.508
.150	-.387	-.199	-.017	.143	.308	-.374	-.197	.007	.170	.336
.250	-.267	-.173	-.047	.066	.192	-.327	-.227	-.082	.049	.186
.350	-.254	-.211	-.119	-.023	.081	-.333	-.298	-.176	-.060	.066
.450	-.253	-.265	-.193	-.117	-.031	-.343	-.380	-.275	-.170	-.062
.550	-.201	-.267	-.219	-.156	-.093	-.298	-.415	-.326	-.240	-.153
.650	-.141	-.206	-.195	-.166	-.122	-.216	-.350	-.319	-.267	-.197
.750	-.076	-.120	-.143	-.145	-.137	-.124	-.110	-.213	-.206	-.189
.850	.010	-.008	-.034	-.055	-.088	-.028	-.002	-.039	-.092	-.137
.900	.049	.030	.007	-.021	-.072	.014	.025	-.023	-.065	-.150
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.704	.909	.781	.495	.102	.802	.879	.731	.459	.173
.025	.360	-.093	-1.107	-1.485	-.887	.249	-.312	-1.027	-1.109	-.387
.075	.154	-.176	-.804	-1.308	-.853	-.046	-.518	-.496	-.340	-.381
.150	-.003	-.245	-.461	-1.130	-.766	-.310	-.714	-.455	-.338	-.381
.250	-.138	-.330	-.403	-.394	-.566	-.456	-.764	-.441	-.343	-.387
.350	-.276	-.462	-.414	-.278	-.516					
.450	-.416	-.595	-.428	-.280	-.435					
.550	-.514	-.685	-.378	-.297	-.434					
.650	-.557	-.569	-.302	-.314	-.449	-.628	-.395	-.398	-.367	-.409
.750	-.233	-.122	-.246	-.328	-.462	-.303	-.303	-.385	-.371	-.409
.850	-.106	-.054	-.189	-.348	-.455	-.184	-.149	-.339	-.360	-.390
.900	-.086	-.009	-.162	-.357	-.459	-.153	-.086	-.316	-.354	-.380
	<i>Right side</i>									
.025	-1.071	-.084	.386	.622	.793	-.748	-.293	.287	.598	.818
.075	-.774	-.192	.133	.343	.535	-.468	-.607	-.076	.219	.464
.150	-.416	-.234	.007	.182	.358	-.457	-.634	-.329	-.072	.170
.250	-.387	-.339	-.146	.016	.185					
.350	-.417	-.472	-.282	-.140	.016					
.450	-.417	-.589	-.431	-.283	-.139					
.550	-.354	-.660	-.524	-.397	-.267					
.650	-.287	-.507	-.548	-.474	-.347	-.368	-.368	-.521	-.649	-.525
.750	-.240	-.125	-.272	-.432	-.334	-.368	-.278	-.345	-.317	-.276
.850	-.176	-.022	-.083	-.143	-.189	-.327	-.148	-.183	-.174	-.174
.900	-.141	.022	-.063	-.142	-.204	-.290	-.078	-.146	-.166	-.183

TABLE III - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(d)  $\alpha = -9.8^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.913	.926	.817	.568	.162	.751	.893	.821	.587	.152
.025	.238	-.095	-.470	-.822	-1.239	.320	-.069	-.893	-1.228	-.960
.075	.104	-.133	-.450	-.740	-.972	.126	-.162	-.681	-1.131	-.948
.150	-.020	-.211	-.417	-.675	-.796	.017	-.191	-.443	-1.057	-.750
.250	-.054	-.178	-.279	-.620	-.683	-.070	-.218	-.295	-.867	-.745
.350	-.096	-.183	-.235	-.274	-.510	-.170	-.285	-.362	-.190	-.649
.450	-.183	-.250	-.257	-.172	-.378	-.295	-.381	-.335	-.147	-.488
.550	-.238	-.295	-.215	-.137	-.286	-.353	-.451	-.317	-.180	-.386
.650	-.272	-.295	-.165	-.112	-.233	-.427	-.506	-.237	-.195	-.333
.750	-.280	-.189	-.093	-.084	-.208	-.449	-.266	-.157	-.179	-.305
.850	-.127	-.033	-.013	-.039	-.161	-.229	.021	-.063	-.124	-.282
.900	-.047	.022	.029	-.004	-.135	-.079	.038	-.007	-.083	-.257
<i>Right side</i>										
.025	-.364	-.023	.294	.535	.787	-.865	-.089	.338	.571	.772
.075	-.479	-.167	.079	.284	.508	-.696	-.183	.115	.313	.522
.150	-.444	-.209	-.021	.139	.330	-.529	-.198	.011	.169	.351
.250	-.312	-.173	-.042	.070	.217	-.293	-.221	-.077	.051	.206
.350	-.258	-.211	-.117	-.031	.094	-.348	-.291	-.171	-.063	.086
.450	-.266	-.259	-.194	-.125	-.022	-.312	-.369	-.277	-.178	-.048
.550	-.213	-.303	-.245	-.183	-.096	-.329	-.453	-.351	-.262	-.152
.650	-.158	-.282	-.259	-.216	-.148	-.236	-.486	-.406	-.325	-.227
.750	-.090	-.191	-.297	-.270	-.219	-.142	-.339	-.431	-.369	-.288
.850	-.003	-.026	-.119	-.156	-.192	-.052	.021	-.247	-.286	-.282
.900	.033	.025	-.049	-.095	-.189	-.016	.033	-.073	-.203	-.340
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.748	.925	.831	.587	.202	.838	.902	.791	.548	.217
.025	.368	-.069	-.977	-1.291	-.836	.296	-.206	-.881	-1.041	-.377
.075	.165	-.154	-.823	-1.156	-.815	-.007	-.439	-.415	-.341	-.380
.150	.017	-.219	-.689	-1.075	-.759	-.268	-.643	-.366	-.333	-.376
.250	-.114	-.298	-.357	-.658	-.619	-.434	-.734	-.349	-.335	-.377
.350	-.256	-.417	-.279	-.276	-.578					
.450	-.407	-.557	-.374	-.222	-.422					
.550	-.529	-.673	-.383	-.281	-.408					
.650	-.604	-.687	-.290	-.313	-.420	-.857	-.378	-.353	-.361	-.398
.750	-.671	-.293	-.270	-.346	-.434	-.735	-.390	-.361	-.363	-.397
.850	-.561	-.038	-.254	-.371	-.438	-.396	-.342	-.347	-.355	-.389
.900	-.284	-.018	-.246	-.385	-.449	-.341	-.298	-.338	-.354	-.382
<i>Right side</i>										
.025	-.967	-.070	.389	.619	.799	-.714	-.230	.317	.615	.847
.075	-.825	-.185	.143	.351	.548	-.399	-.535	-.041	.240	.498
.150	-.660	-.222	.023	.194	.381	-.375	-.582	-.299	-.043	.208
.250	-.341	-.309	-.117	.034	.212					
.350	-.296	-.438	-.262	-.120	.048					
.450	-.380	-.566	-.395	-.276	-.111					
.550	-.374	-.653	-.538	-.387	-.247					
.650	-.299	-.690	-.595	-.492	-.346	-.371	-.439	-.799	-.731	-.570
.750	-.287	-.347	-.654	-.561	-.462	-.372	-.406	-.681	-.811	-.655
.850	-.260	-.013	-.607	-.589	-.478	-.349	-.318	-.435	-.617	-.613
.900	-.242	.014	-.291	-.575	-.494	-.333	-.258	-.358	-.387	-.412

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(e)  $\alpha = -9.8^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.927	.940	.831	.573	.191	.746	.898	.829	.593	.185
.025	.270	-.095	-.468	-.876	-1.215	.341	-.066	-.866	-1.184	-.797
.075	.126	-.135	-.490	-.749	-.969	.142	-.172	-.728	-1.094	-.798
.150	-.006	-.222	-.480	-.715	-.830	.031	-.199	-.578	-1.052	-.806
.250	-.043	-.188	-.513	-.722	-.661	-.058	-.218	-.536	-.965	-.772
.350	-.090	-.192	-.291	-.479	-.545	-.160	-.285	-.388	-.653	-.642
.450	-.177	-.256	-.279	-.258	-.439	-.273	-.372	-.365	-.149	-.510
.550	-.252	-.315	-.236	-.120	-.353	-.355	-.460	-.258	-.112	-.442
.650	-.279	-.334	-.186	-.105	-.307	-.429	-.524	-.283	-.161	-.417
.750	-.338	-.332	-.130	-.094	-.282	-.459	-.535	-.212	-.191	-.410
.850	-.246	-.075	-.046	-.067	-.245	-.450	-.067	-.123	-.171	-.403
.900	-.116	.006	-.004	-.040	-.212	-.271	.031	-.060	-.140	-.361
	<i>Right side</i>									
.025	-.379	-.028	.292	.555	.795	-.861	-.096	.329	.579	.769
.075	-.528	-.177	.076	.299	.506	-.768	-.197	.109	.322	.516
.150	-.490	-.232	-.028	.152	.333	-.619	-.211	.001	.178	.348
.250	-.470	-.180	-.051	.072	.211	-.495	-.231	-.081	.058	.207
.350	-.293	-.216	-.126	-.027	.090	-.343	-.298	-.176	-.055	.080
.450	-.273	-.274	-.207	-.126	-.026	-.270	-.368	-.277	-.177	-.054
.550	-.206	-.324	-.278	-.194	-.106	-.288	-.463	-.375	-.258	-.157
.650	-.175	-.311	-.279	-.226	-.164	-.276	-.519	-.420	-.335	-.236
.750	-.118	-.344	-.362	-.322	-.258	-.178	-.489	-.456	-.377	-.309
.850	-.025	-.067	-.281	-.263	-.235	-.087	-.043	-.471	-.386	-.309
.900	.010	.006	-.177	-.242	-.309	-.054	.031	-.442	-.412	-.385
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.749	.930	.844	.600	.228	.845	.916	.810	.568	.245
.025	.385	-.058	-.941	-1.230	-.866	.327	-.211	-.891	-1.027	-.458
.075	.183	-.154	-.811	-1.114	-.865	.026	-.413	-.416	-.374	-.435
.150	.034	-.223	-.764	-1.074	-.789	-.238	-.631	-.372	-.364	-.418
.250	-.098	-.292	-.675	-.970	-.579	-.411	-.737	-.363	-.368	-.418
.350	-.235	-.408	-.321	-.428	-.471					
.450	-.384	-.545	-.240	-.279	-.451					
.550	-.514	-.667	-.340	-.294	-.462					
.650	-.590	-.730	-.340	-.349	-.476	-.829	-.395	-.374	-.382	-.431
.750	-.656	-.644	-.341	-.382	-.490	-.913	-.408	-.389	-.386	-.431
.850	-.648	-.126	-.350	-.403	-.480	-.684	-.404	-.382	-.383	-.421
.900	-.671	-.050	-.353	-.420	-.485	-.527	-.379	-.376	-.385	-.413
	<i>Right side</i>									
.025	-.953	-.072	.388	.624	.799	-.711	-.214	.330	.631	.850
.075	-.832	-.186	.143	.360	.549	-.394	-.518	-.030	.262	.503
.150	-.748	-.224	.023	.204	.373	-.375	-.589	-.275	-.021	.208
.250	-.681	-.313	-.118	.046	.212					
.350	-.151	-.434	-.260	-.114	.047					
.450	-.272	-.566	-.396	-.262	-.110					
.550	-.358	-.666	-.539	-.379	-.243					
.650	-.339	-.742	-.606	-.475	-.362	-.390	-.484	-.834	-.717	-.574
.750	-.343	-.665	-.665	-.557	-.458	-.391	-.456	-.877	-.797	-.655
.850	-.337	-.133	-.663	-.581	-.497	-.375	-.400	-.674	-.788	-.658
.900	-.326	-.037	-.691	-.612	-.530	-.359	-.351	-.564	-.699	-.619

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(f)  $\alpha = 0^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.606	.675	.344	-.455	-1.251	.271	.749	.314	-.393	-1.072
.025	.287	-.090	-.546	-1.109	-1.849	.354	-.086	-.649	-1.040	-1.233
.075	.142	-.111	-.378	-.665	-1.435	.170	-.125	-.447	-.920	-.920
.150	.050	-.136	-.318	-.464	-.874	.075	-.136	-.341	-.574	-.893
.250	.029	-.100	-.212	-.284	-.444	.006	-.146	-.295	-.409	-.817
.350	-.017	-.113	-.189	-.231	-.336	-.052	-.178	-.290	-.319	-.707
.450	-.061	-.125	-.178	-.195	-.295	-.089	-.194	-.270	-.266	-.587
.550	-.052	-.109	-.141	-.151	-.293	-.089	-.159	-.208	-.220	-.470
.650	-.038	-.077	-.090	-.109	-.217	-.066	-.123	-.155	-.176	-.392
.750	-.015	-.026	-.031	-.059	-.148	-.029	-.061	-.079	-.123	-.313
.850	.029	.025	.025	.010	-.072	.024	.013	.006	-.052	-.233
.900	.052	.059	.057	.020	-.028	.059	.052	.048	-.008	-.182
	<i>Right side</i>									
.025	-.482	-.035	.321	.612	.831	-.648	-.061	.390	.610	.744
.075	-.394	-.118	.130	.354	.556	-.447	-.116	.169	.382	.539
.150	-.315	-.129	.050	.218	.381	-.350	-.132	.075	.239	.390
.250	-.228	-.104	.020	.156	.275	-.292	-.146	.013	.137	.268
.350	-.202	-.118	-.024	.075	.174	-.274	-.159	-.035	.068	.183
.450	-.184	-.136	-.067	.015	.098	-.200	-.136	-.084	-.022	.049
.550	-.140	-.104	-.054	.008	.073	-.211	-.150	-.077	-.015	.052
.650	-.080	-.070	-.033	.006	.061	-.147	-.111	-.054	-.015	.034
.750	-.031	-.021	-.001	.024	.048	-.070	-.040	-.005	.004	.031
.850	.043	.041	.050	.057	.073	.024	.036	.059	.047	.031
.900	.066	.066	.064	.066	.068	.052	.052	.059	.047	.011
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.163	.751	.220	-.324	-.654	.394	.779	.036	-.248	-.509
.025	.403	-.109	-.959	-.987	-.720	.495	-.313	-1.357	-.701	-.484
.075	.209	-.139	-.546	-.936	-.743	.221	-.316	-1.152	-.367	
.150	.091	-.164	-.433	-.840	-.775	.003	-.334	-1.033	-.354	
.250	.010	-.182	-.378	-.598	-.707	-.103	-.329	-.605	-.356	-.449
.350	-.054	-.215	-.364	-.377	-.546					
.450	-.105	-.233	-.336	-.275	-.511					
.550	-.119	-.212	-.281	-.261	-.504					
.650	-.089	-.159	-.198	-.277	-.493	-.131	.180	-.185	-.397	-.438
.750	-.038	-.077	-.111	-.280	-.461	-.054	.079	-.093	-.390	-.401
.850	.017	-.003	-.024	-.252	-.408	.022	.025	-.003	-.351	-.348
.900	.031	.027	.015	-.236	-.385	.052	.064	.029	-.328	-.320
	<i>Right side</i>									
.025	-.805	-.051	.441	.640	.742	-1.264	.164	.562	.746	.910
.075	-.532	-.129	.208	.409	.558	-.897	.281	.215	.421	.631
.150	-.396	-.136	.103	.271	.418	-1.038	.297	.022	.199	.390
.250	-.375	-.178	.020	.163	.298					
.350	-.371	-.208	-.056	.057	.192					
.450	-.334	-.219	-.097	-.015	.100					
.550	-.278	-.205	-.111	-.054	.029					
.650	-.195	-.150	-.088	-.056	-.001	-.114	.129	.111	-.112	-.086
.750	-.103	-.074	-.038	-.047	-.024	-.077	.072	-.056	-.130	-.111
.850	.006	.027	.041	-.017	-.042	.017	.041	.031	-.105	-.109
.900	.047	.050	.061	-.022	-.081	.047	.080	.061	-.118	-.136

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(g)  $\alpha = 0^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.726	.680	.546	.190	-.264	.496	.781	.549	.075	-.348
.025	.295	-.059	-.525	-1.131	-.945	.353	-.068	-.932	-1.224	-.670
.075	.158	-.092	-.367	-.851	-.840	.175	-.121	-.442	-1.046	-.627
.150	.058	-.132	-.324	-.471	-.746	.082	-.140	-.336	-.851	-.653
.250	.035	-.098	-.212	-.282	-.620	.005	-.156	-.286	-.518	-.653
.350	-.018	-.123	-.196	-.229	-.519	-.061	-.202	-.289	-.312	-.591
.450	-.070	-.149	-.188	-.197	-.463	-.122	-.231	-.283	-.236	-.513
.550	-.088	-.123	-.145	-.147	-.372	-.117	-.194	-.225	-.188	-.430
.650	-.059	-.091	-.094	-.106	-.290	-.094	-.146	-.159	-.159	-.352
.750	-.024	-.033	-.030	-.063	-.206	-.044	-.061	-.071	-.116	-.279
.850	.027	.034	.036	-.002	-.121	.026	.028	.016	-.051	-.228
.900	.061	.070	.076	.031	-.073	.067	.073	.062	-.004	-.199
<i>Right side</i>										
.025	-.411	.006	.340	.606	.829	-.976	-.042	.386	.617	.754
.075	-.390	-.095	.149	.360	.561	-.429	-.112	.177	.384	.549
.150	-.314	-.121	.060	.228	.402	-.335	-.123	.085	.249	.403
.250	-.216	-.098	.027	.158	.295	-.277	-.135	.014	.148	.286
.350	-.199	-.127	-.026	.075	.192	-.269	-.171	-.048	.067	.195
.450	-.186	-.146	-.073	.008	.104	-.219	-.168	-.094	-.016	.084
.550	-.137	-.120	-.068	-.009	.067	-.222	-.185	-.117	-.043	.047
.650	-.075	-.080	-.050	-.010	.043	-.146	-.124	-.088	-.043	.017
.750	-.021	-.024	-.018	.001	.028	-.053	-.045	-.032	-.021	.003
.850	.053	.047	.042	.045	.043	.040	.049	.045	.026	.000
.900	.090	.076	.060	.057	.031	.079	.079	.063	.026	-.033
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.452	.781	.497	.125	-.380	.677	.827	.537	.148	-.202
.025	.400	-.086	-1.070	-.714	-.735	.469	-.298	-.944	-.567	-.381
.075	.218	-.127	-.519	-.701	-.746	.192	-.383	-.607	-.291	-.372
.150	.104	-.165	-.393	-.664	-.747	-.038	-.393	-.577	-.288	-.366
.250	.008	-.206	-.341	-.562	-.738	-.148	-.375	-.560	-.292	-.381
.350	-.078	-.261	-.349	-.418	-.491					
.450	-.155	-.299	-.346	-.303	-.434					
.550	-.178	-.272	-.295	-.254	-.437					
.650	-.145	-.190	-.211	-.250	-.454	-.190	-.171	-.281	-.342	-.412
.750	-.068	-.071	-.125	-.260	-.456	-.090	-.036	-.232	-.347	-.393
.850	.002	.012	-.041	-.250	-.430	-.020	.067	-.169	-.333	-.345
.900	.026	.049	-.003	-.238	-.415	.003	.105	-.133	-.315	-.323
<i>Right side</i>										
.025	-1.055	-.030	.437	.641	.762	-1.073	-.150	.510	.770	.950
.075	-.487	-.114	.216	.419	.585	-.524	-.339	.178	.442	.663
.150	-.358	-.130	.112	.290	.450	-.577	-.343	-.015	.202	.420
.250	-.347	-.190	.011	.170	.326					
.350	-.361	-.243	-.073	.057	.204					
.450	-.345	-.279	-.143	-.030	.095					
.550	-.283	-.257	-.171	-.091	.017					
.650	-.193	-.174	-.140	-.109	-.033	-.050	-.091	-.122	-.141	-.123
.750	-.103	-.071	-.075	-.091	-.068	-.193	-.030	-.102	-.165	-.149
.850	.000	.049	.022	-.046	-.068	-.135	.088	-.018	-.127	-.133
.900	.046	.088	.051	-.053	-.108	-.091	.123	-.003	-.138	-.155

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(h)  $\alpha = 0^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.748	.685	.596	.304	-.114	.548	.795	.609	.190	-.225
.025	.291	-.052	-.515	-1.094	-.928	.347	-.064	-1.079	-1.548	-.582
.075	.154	-.090	-.374	-.862	-.776	.169	-.122	-.408	-1.364	-.588
.150	.052	-.135	-.334	-.617	-.698	.072	-.142	-.339	-.666	-.614
.250	.034	-.106	-.218	-.270	-.581	-.002	-.159	-.288	-.306	-.624
.350	-.025	-.133	-.195	-.221	-.509	-.074	-.215	-.288	-.293	-.581
.450	-.084	-.167	-.195	-.202	-.473	-.140	-.260	-.283	-.258	-.517
.550	-.086	-.143	-.151	-.149	-.404	-.147	-.230	-.226	-.201	-.445
.650	-.081	-.110	-.098	-.112	-.330	-.129	-.172	-.162	-.162	-.379
.750	-.043	-.046	-.029	-.070	-.253	-.070	-.073	-.083	-.117	-.311
.850	.015	.033	.036	-.012	-.168	.012	.030	.007	-.091	-.253
.900	.045	.073	.071	.019	-.117	.051	.076	.059	-.014	-.220
	<i>Right side</i>									
.025	-.387	.011	.336	.603	.837	-.982	-.042	.380	.605	.766
.075	-.393	-.099	.143	.354	.572	-.429	-.116	.170	.370	.557
.150	-.323	-.129	.059	.225	.408	-.334	-.133	.074	.240	.408
.250	-.224	-.109	.022	.151	.302	-.275	-.150	.002	.141	.290
.350	-.202	-.137	-.040	.065	.192	-.270	-.192	-.064	.058	.196
.450	-.193	-.166	-.096	-.012	.096	-.241	-.208	-.115	-.035	.084
.550	-.142	-.140	-.095	-.030	.062	-.224	-.216	-.147	-.073	.033
.650	-.083	-.097	-.075	-.034	.031	-.149	-.157	-.121	-.077	-.006
.750	-.020	-.037	-.040	-.033	.010	-.060	-.054	-.057	-.053	-.020
.850	.051	.046	.033	.024	.025	.034	.050	.028	-.001	-.024
.900	.082	.077	.051	.034	.010	.069	.081	.052	.003	-.065
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.505	.795	.563	.144	-.267	.730	.855	.619	.209	-.146
.025	.396	-.076	-1.155	-.962	-.696	.442	-.272	-.889	-.522	-.404
.075	.215	-.125	-.464	-.910	-.703	.177	-.425	-.493	-.293	-.384
.150	.095	-.165	-.374	-.770	-.706	-.060	-.429	-.437	-.293	-.368
.250	-.004	-.215	-.313	-.550	-.727	-.175	-.389	-.429	-.298	-.377
.350	-.101	-.290	-.323	-.364	-.502					
.450	-.188	-.348	-.326	-.267	-.435					
.550	-.232	-.345	-.283	-.242	-.432					
.650	-.195	-.230	-.220	-.251	-.449	-.235	-.155	-.302	-.336	-.405
.750	-.107	-.073	-.147	-.271	-.450	-.127	-.024	-.283	-.347	-.386
.850	-.022	.024	-.080	-.274	-.432	-.057	.073	-.237	-.339	-.355
.900	.001	.061	-.043	-.264	-.423	-.035	.113	-.197	-.330	-.337
	<i>Right side</i>									
.025	-1.145	-.030	.433	.644	.771	-1.012	-.143	.501	.779	.962
.075	-.436	-.112	.212	.419	.588	-.413	-.395	.161	.443	.573
.150	-.334	-.133	.110	.288	.457	-.435	-.382	-.039	.196	.420
.250	-.317	-.203	.005	.164	.332					
.350	-.337	-.272	-.096	.045	.207					
.450	-.328	-.326	-.180	-.060	.088					
.550	-.278	-.318	-.227	-.139	-.010					
.650	-.206	-.200	-.193	-.155	-.065	-.168	-.139	-.157	-.183	-.154
.750	-.130	-.076	-.112	-.130	-.102	-.255	-.022	-.135	-.205	-.181
.850	-.038	.056	-.006	-.073	-.091	-.208	.090	-.054	-.149	-.151
.900	.006	.097	.025	-.081	-.127	-.169	.129	-.044	-.165	-.175

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(1)  $\alpha = 0^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.780	.696	.649	.412	.032	.607	.810	.669	.325	-.110
.025	.300	-.042	-.458	-.993	-.922	.348	-.062	-.979	-1.344	-.541
.075	.165	-.079	-.371	-.778	-.739	.173	-.125	-.436	-1.216	-.542
.150	.061	-.140	-.341	-.648	-.647	.079	-.137	-.319	-1.092	-.556
.250	.036	-.105	-.213	-.396	-.526	.005	-.156	-.272	-.236	-.566
.350	-.027	-.136	-.193	-.201	-.470	-.077	-.227	-.274	-.245	-.553
.450	-.091	-.182	-.195	-.182	-.459	-.160	-.291	-.281	-.270	-.507
.550	-.101	-.171	-.146	-.146	-.416	-.183	-.295	-.224	-.198	-.459
.650	-.114	-.141	-.094	-.111	-.363	-.192	-.227	-.158	-.157	-.414
.750	-.075	-.063	-.024	-.067	-.305	-.124	-.087	-.078	-.112	-.360
.850	.001	.031	.047	-.015	-.234	-.007	.035	.011	-.052	-.319
.900	.043	.081	.079	.031	-.190	.047	.090	.064	-.011	-.283
	<i>Right side</i>									
.025	-.331	.031	.348	.616	.846	-.921	-.028	.377	.610	.771
.075	-.398	-.086	.161	.371	.580	-.554	-.110	.181	.383	.566
.150	-.344	-.120	.068	.241	.416	-.327	-.126	.080	.250	.414
.250	-.219	-.106	.032	.163	.307	-.260	-.151	.008	.151	.294
.350	-.200	-.139	-.032	.074	.194	-.258	-.200	-.062	.066	.199
.450	-.195	-.182	-.099	-.011	.094	-.239	-.235	-.134	-.032	.082
.550	-.141	-.165	-.111	-.042	.045	-.219	-.268	-.181	-.086	.016
.650	-.082	-.124	-.100	-.056	.004	-.144	-.207	-.179	-.112	-.035
.750	-.020	-.051	-.071	-.063	-.031	-.055	-.066	-.110	-.102	-.064
.850	.060	.043	.013	-.003	-.017	.035	.057	.007	-.038	-.070
.900	.087	.081	.043	.016	-.044	.071	.086	.035	-.020	-.110
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.567	.812	.629	.261	-.117	.781	.884	.687	.313	-.171
.025	.399	-.066	-1.067	-.874	-.594	.458	-.215	-.895	-.493	-.609
.075	.222	-.114	-.855	-.821	-.598	.186	-.427	-.400	-.288	-.394
.150	.103	-.157	-.305	-.750	-.609	-.059	-.484	-.328	-.283	-.371
.250	.001	-.210	-.245	-.588	-.605	-.157	-.448	-.321	-.281	-.374
.350	-.104	-.322	-.278	-.422	-.580					
.450	-.215	-.416	-.305	-.305	-.544					
.550	-.303	-.432	-.268	-.222	-.454					
.650	-.326	-.371	-.212	-.226	-.426	-.539	-.183	-.287	-.308	-.410
.750	-.205	-.054	-.155	-.260	-.424	-.142	-.058	-.283	-.321	-.404
.850	-.037	.040	-.104	-.281	-.415	-.066	.039	-.250	-.313	-.386
.900	-.010	.078	-.072	-.284	-.412	-.054	.081	-.229	-.311	-.370
	<i>Right side</i>									
.025	-1.047	-.009	.434	.651	.782	-.953	-.093	.505	.796	.977
.075	-.787	-.097	.225	.430	.600	-.317	-.406	.171	.459	.683
.150	-.272	-.120	.124	.304	.464	-.313	-.413	-.032	.215	.423
.250	-.259	-.195	.011	.181	.337					
.350	-.289	-.297	-.092	.054	.206					
.450	-.302	-.377	-.201	-.062	.079					
.550	-.264	-.410	-.287	-.165	-.034					
.650	-.199	-.343	-.324	-.224	-.121	-.181	-.183	-.317	-.411	-.297
.750	-.146	-.077	-.236	-.265	-.174	-.270	-.051	-.158	-.264	-.264
.850	-.063	.066	-.025	-.103	-.135	-.239	.062	-.062	-.149	-.202
.900	-.033	.109	.004	-.100	-.165	-.211	.105	-.055	-.158	-.228

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(j)  $\alpha = 0^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.788	.694	.660	.452	.076	.627	.813	.681	.368	-.074
.025	.298	-.040	-.448	-.968	-.908	.345	-.060	-.958	-1.279	-.523
.075	.167	-.084	-.377	-.746	-.726	.171	-.123	-.669	-1.162	-.531
.150	.061	-.140	-.351	-.636	-.637	.075	-.144	-.386	-1.063	-.544
.250	.034	-.107	-.224	-.512	-.513	.005	-.157	-.265	-.569	-.552
.350	-.032	-.140	-.200	-.270	-.452	-.075	-.231	-.293	-.143	-.541
.450	-.107	-.191	-.208	-.166	-.451	-.155	-.299	-.286	-.199	-.502
.550	-.117	-.186	-.156	-.126	-.410	-.204	-.324	-.249	-.179	-.451
.650	-.142	-.172	-.105	-.106	-.366	-.237	-.302	-.159	-.160	-.405
.750	-.119	-.076	-.029	-.065	-.312	-.200	-.099	-.076	-.126	-.368
.850	-.029	.028	.045	-.009	-.244	-.050	.037	.014	-.065	-.320
.900	.025	.074	.079	.022	-.206	.024	.093	.067	-.029	-.285
<i>Right side</i>										
.025	-.319	.035	.354	.617	.845	-.895	-.023	.381	.613	.778
.075	-.407	-.085	.162	.373	.586	-.636	-.107	.179	.385	.567
.150	-.369	-.120	.072	.242	.422	-.372	-.123	.087	.254	.418
.250	-.242	-.109	.032	.168	.313	-.254	-.147	.013	.151	.301
.350	-.216	-.147	-.034	.076	.204	-.265	-.198	-.061	.066	.203
.450	-.207	-.191	-.100	-.014	.098	-.246	-.249	-.132	-.037	.093
.550	-.153	-.189	-.117	-.047	.051	-.237	-.296	-.185	-.097	.018
.650	-.092	-.151	-.116	-.074	-.001	-.153	-.271	-.198	-.135	-.034
.750	-.024	-.069	-.095	-.094	-.039	-.063	-.080	-.141	-.141	-.075
.850	.051	.040	.004	-.034	-.030	.026	.053	-.012	-.091	-.089
.900	.082	.074	.034	-.017	-.062	.062	.090	.029	-.064	-.133
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.584	.821	.640	.305	-.081	.801	.894	.701	.359	-.131
.025	.402	-.056	-1.075	-1.122	-.579	.466	-.197	-.950	-.483	-.611
.075	.228	-.107	-.864	-1.018	-.582	.196	-.399	-.403	-.285	-.389
.150	.111	-.147	-.287	-.839	-.594	-.049	-.479	-.449	-.281	-.362
.250	.010	-.198	-.233	-.542	-.591	-.149	-.463	-.391	-.284	-.367
.350	-.099	-.311	-.287	-.372	-.581					
.450	-.217	-.420	-.352	-.285	-.552					
.550	-.308	-.484	-.271	-.217	-.450					
.650	-.387	-.449	-.210	-.239	-.418	-.642	-.216	-.274	-.307	-.403
.750	-.368	-.052	.150	-.270	-.412	-.378	-.123	-.262	-.315	-.401
.850	-.082	.049	-.095	-.288	-.405	-.116	-.026	-.235	-.305	-.384
.900	-.020	.085	-.065	-.296	-.408	-.079	.019	-.210	-.305	-.374
<i>Right side</i>										
.025	-1.026	-.002	.439	.652	.789	-.939	-.068	.522	.801	.980
.075	-.810	-.090	.228	.429	.604	-.287	-.377	.184	.460	.687
.150	-.231	-.109	.130	.309	.469	-.285	-.417	-.026	.213	.427
.250	-.242	-.193	.018	.185	.343					
.350	-.266	-.295	-.090	.058	.211					
.450	-.303	-.388	-.203	-.061	.080					
.550	-.262	-.437	-.285	-.170	-.033					
.650	-.200	-.445	-.347	-.239	-.125	-.283	-.249	-.513	-.527	-.382
.750	-.157	-.082	-.329	-.322	-.218	-.274	-.164	-.248	-.510	-.340
.850	-.087	.066	-.040	-.233	-.164	-.246	-.036	-.067	-.266	-.239
.900	-.051	.110	.005	-.167	-.195	-.215	.027	-.046	-.173	-.261



TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(k)  $\alpha = 9.4^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.412	.795	.434	-.723	-1.227	.143	.689	.130	-.757	-1.197
.025	-.011	-.124	-.240	-.711	-1.957	.353	-.082	-.749	-1.276	-.950
.075	.034	-.122	-.321	-.521	-1.114	.180	-.115	-.465	-1.170	-.932
.150	.000	-.145	-.312	-.426	-.625	.087	-.122	-.339	-.841	-.886
.250	.003	-.093	-.226	-.299	-.411	.037	-.127	-.283	-.351	-.835
.350	-.020	-.104	-.211	-.253	-.344	-.011	-.142	-.271	-.287	-.708
.450	-.038	-.115	-.202	-.231	-.335	-.050	-.158	-.249	-.276	-.565
.550	-.029	-.095	-.161	-.190	-.277	-.041	-.127	-.195	-.231	-.457
.650	-.016	-.062	-.116	-.165	-.215	-.027	-.091	-.141	-.192	-.379
.750	.007	-.023	-.064	-.095	-.162	-.004	-.037	-.073	-.129	-.319
.850	.037	.028	-.001	-.027	-.095	.041	.026	.010	-.047	-.236
.900	.057	.058	.033	.003	-.049	.062	.058	.046	-.002	-.183
	<i>Right side</i>									
.025	-.172	-.039	.015	.513	.794	-.677	-.046	.402	.606	.718
.075	-.316	-.127	.026	.332	.536	-.438	-.104	.195	.393	.545
.150	-.295	-.131	.010	.232	.388	-.334	-.111	.100	.264	.409
.250	-.227	-.086	.001	.171	.294	-.268	-.118	.046	.166	.303
.350	-.213	-.107	-.019	.084	.190	-.247	-.129	.006	.112	.220
.450	-.202	-.122	-.044	.039	.112	-.229	-.140	.026	.053	.140
.550	-.150	-.089	-.026	.023	.080	-.184	-.115	-.042	.023	.089
.650	-.104	-.062	-.008	.021	.059	-.127	-.084	-.012	.019	.061
.750	-.054	-.021	.015	.028	.047	-.054	-.021	.015	.028	.041
.850	.016	.040	.055	.057	.059	.030	.046	.064	.059	.043
.900	.043	.062	.064	.066	.054	.053	.060	.064	.057	.011
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.003	.660	-.042	-.453	-.874	.407	.810	-.213	-.308	-.824
.025	.418	-.084	-1.057	-.766	-.609	.587	-.174	-1.199	-.464	-.441
.075	.246	-.104	-.551	-.736	-.612	.339	-.181	-1.017	-.478	-.413
.150	.148	-.111	-.395	-.666	-.595	.139	-.203	-.859	-.471	-.423
.250	.082	-.122	-.328	-.557	-.565	.053	-.167	-.515	-.464	-.432
.350	.021	-.147	-.303	-.467	-.542					
.450	-.022	-.163	-.280	-.405	-.515					
.550	-.054	-.151	-.238	-.369	-.496					
.650	-.036	-.107	-.168	-.335	-.469	-.075	-.131	-.161	-.378	-.402
.750	-.004	-.046	-.089	-.285	-.446	-.025	-.053	-.069	-.331	-.379
.850	.021	.010	-.017	-.219	-.411	.018	.019	-.001	-.276	-.349
.900	.041	.037	.024	-.190	-.397	.046	.058	.042	-.249	-.330
	<i>Right side</i>									
.025	-.945	-.019	.443	.620	.704	-1.097	-.071	.679	.855	.997
.075	-.500	-.075	.251	.440	.580	-.968	-.124	.350	.545	.739
.150	-.350	-.082	.166	.320	.462	-.838	.115	.195	.357	.527
.250	-.311	-.107	.096	.230	.368					
.350	-.297	-.129	.033	.143	.259					
.450	-.263	-.145	-.012	.078	.170					
.550	-.218	-.140	-.033	.023	.091					
.650	-.150	-.100	-.024	-.002	.031	-.116	-.086	-.021	-.029	-.033
.750	-.072	-.041	.003	-.002	-.015	-.059	-.039	-.010	-.065	-.114
.850	.018	.040	.060	.021	-.052	.028	.042	.048	-.056	-.146
.900	.055	.064	.069	.007	-.086	.055	.062	.060	-.083	-.181

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(i)  $\alpha = 9.6^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.379	.864	.433	-.404	-.214	.417	.740	.433	-.133	-.707
.025	-.026	-.102	-.153	-.566	-1.347	.360	.076	-.983	-1.143	-.779
.075	.038	-.117	-.313	-.519	-1.009	.187	.120	-.488	-1.078	-.793
.150	.009	-.154	-.351	-.457	-.719	.102	.120	-.354	-.898	-.759
.250	.006	-.099	-.244	-.315	-.467	.042	.130	-.294	-.514	-.719
.350	-.020	-.106	-.224	-.263	-.382	-.022	.162	-.293	-.316	-.651
.450	-.047	-.132	-.221	-.238	-.343	-.061	.183	-.279	-.260	-.561
.550	-.041	-.108	-.179	-.196	-.296	-.061	-.153	-.223	-.229	-.472
.650	-.032	-.078	-.133	-.171	-.244	-.049	-.117	-.160	-.206	-.412
.750	-.006	-.034	-.070	-.125	-.189	-.015	-.046	-.075	-.163	-.355
.850	.039	.032	.000	-.045	-.118	.042	.035	.017	-.084	-.287
.900	.067	.068	.038	-.010	-.069	.076	.080	.062	-.032	-.230
	<i>Right side</i>									
.025	-.092	-.007	.001	.425	.800	-1.070	-.028	.411	.622	.747
.075	-.325	-.117	.047	.334	.548	-.444	-.097	.207	.409	.571
.150	-.337	-.126	.030	.246	.399	-.345	.106	.117	.285	.436
.250	-.249	-.084	.010	.189	.306	-.288	.120	.056	.193	.327
.350	-.237	-.106	-.020	.109	.201	-.276	.139	.010	.126	.243
.450	-.235	-.133	-.044	.033	.112	-.273	.159	-.037	.051	.154
.550	-.180	-.102	-.034	.020	.088	-.226	.148	-.051	.011	.092
.650	-.124	-.064	-.019	.002	.052	-.151	.111	-.034	-.004	.055
.750	-.067	-.025	.007	.008	.038	-.058	.031	.001	.004	.032
.850	.013	.047	.056	.045	.048	.033	.052	.062	.033	.015
.900	.047	.070	.068	.050	.034	.070	.068	.073	.023	-.018
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.305	.706	.300	-.075	-.654	.661	.876	.367	.063	-.587
.025	.427	-.085	-1.090	-.633	-.555	.617	-.228	-.941	-.353	-.392
.075	.259	-.097	-.651	-.667	-.571	.326	.246	-.744	-.375	-.393
.150	.163	-.114	-.394	-.641	-.558	.116	.264	-.660	-.416	-.395
.250	.093	-.132	-.330	-.525	-.551	.029	.210	-.605	-.407	-.402
.350	.021	-.177	-.336	-.378	-.518					
.450	-.047	-.214	-.328	-.355	-.488					
.550	-.089	-.207	-.276	-.364	-.464					
.650	-.070	-.150	-.189	-.364	-.447	-.116	.148	-.206	-.370	-.395
.750	-.025	-.060	-.098	-.343	-.435	-.052	.037	-.147	-.352	-.373
.850	.021	.014	-.014	-.294		.007	.050	-.075	-.312	-.343
.900	.052	.053	.026	-.266	-.395	.038	.089	-.044	-.290	-.324
	<i>Right side</i>									
.025	-1.035	.005	.472	.650	.759	-.900	.085	.669	.886	1.063
.075	-.543	-.066	.276	.469	.630	-.804	.168	.350	.570	.791
.150	-.352	-.069	.186	.354	.518	-.801	.141	.181	.376	.573
.250	-.349	-.114	.110	.256	.413					
.350	-.366	-.157	.035	.158	.301					
.450	-.343	-.187	-.032	.069	.194					
.550	-.273	-.187	-.072	-.001	.100					
.650	-.180	-.141	-.060	-.039	.031	-.139	.091	-.061	-.078	-.035
.750	-.083	-.052	-.020	-.052	-.032	-.072	.033	-.041	-.120	-.143
.850	.024	.047	.050	-.030	-.065	-.003	.062	.020	-.114	-.197
.900	.070	.085	.071	-.048	-.118	.036	.098	.027	-.128	-.247

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_1 = 0^\circ$  - Continued

(m)  $\alpha = 9.7^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.368	.881	.433	-.292		.487	.751	.508	-.003	
.025	-.046	-.099	-.094	-.515		.350	-.091	-1.149	-1.220	
.075	.035	-.122	-.297	-.524		.183	-.130	-.442	-1.046	
.150	.006	-.167	-.362	-.505		.096	-.130	-.345	-.834	
.250	.006	-.112	-.242	-.328		.038	-.140	-.287	-.549	
.350	-.029	-.122	-.220	-.272		-.031	-.179	-.280	-.345	
.450	-.059	-.147	-.223	-.245		-.080	-.213	-.274	-.268	
.550	-.052	-.127	-.173	-.200		-.080	-.184	-.222	-.233	
.650	-.048	-.101	-.134	-.176		-.076	-.143	-.162	-.216	
.750	-.019	-.049	-.075	-.140		-.038	-.065	-.085	-.180	
.850	.036	.029	.001	-.066		.032	.027	.010	-.109	
.900	.062	.067	.044	-.022		.066	.077	.062	-.063	
	<i>Right side</i>									
.025	-.041	-.004	-.019	.362		-1.023	-.021	.403	.624	
.075	-.313	-.119	.048	.318		-.422	-.092	.204	.411	
.150	-.341	-.123	.034	.246		-.330	-.106	.118	.285	
.250	-.244	-.089	.011	.189		-.273	-.122	.058	.192	
.350	-.234	-.116	-.021	.112		-.268	-.147	.010	.123	
.450	-.236	-.147	-.053	.033		-.268	-.173	-.048	.046	
.550	-.182	-.113	-.044	.016		-.229	-.170	-.067	.003	
.650	-.123	-.084	-.031	-.010		-.155	-.126	-.057	-.020	
.750	-.073	-.042	-.007	-.007		-.063	-.040	-.019	-.019	
.850	.015	.038	.052	.031		.029	.051	.050	.011	
.900	.044	.070	.065	.037		.066	.073	.062	-.007	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.377	.729	.383	-.019		.734	.899	.489	.140	
.025	.417	-.086	-1.144	-.647		.593	-.286	-.831	-.339	
.075	.257	-.103	-.619	-.678		.308	-.282	-.591	-.356	
.150	.160	-.125	-.372	-.658		.095	-.299	-.490	-.405	
.250	.085	-.147	-.300	-.532		.015	-.230	-.469	-.395	
.350	.009	-.208	-.306	-.362						
.450	-.072	-.247	-.311	-.348						
.550	-.122	-.244	-.273	-.371						
.650	-.107	-.170	-.200	-.379		-.170	-.154	-.266	-.371	
.750	-.058	-.066	-.122	-.369		-.096	-.035	-.227	-.356	
.850	-.001	.020	-.047	-.333		-.038	.060	-.172	-.322	
.900	.025	.061	-.006	-.303		-.012	.097	-.139	-.303	
	<i>Right side</i>									
.025	-1.196	.021	.473	.657		-.666	-.085	.659	.894	
.075	-.454	-.055	.277	.469		-.586	-.190	.337	.572	
.150	-.314	-.062	.190	.359		-.560	-.150	.173	.370	
.250	-.301	-.112	.108	.266						
.350	-.331	-.173	.024	.153						
.450	-.331	-.211	-.051	.057						
.550	-.276	-.221	-.104	-.023						
.650	-.192	-.160	-.102	-.062		-.240	-.102	-.101	-.116	
.750	-.107	-.061	-.057	-.089		-.196	-.034	-.084	-.159	
.850	-.004	.050	.027	-.063		-.135	.070	-.024	-.142	
.900	.044	.087	.052	-.084		-.086	.105	-.019	-.170	

TABLE III - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(n)  $\alpha = 9.7^\circ$ ;  $M = 0.90$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.323	.927	.424	-.203		.549	.761	.572	.144	
.025	-.069	-.077	-.029	-.443		.353	-.081	-1.046	-1.302	
.075	.039	-.106	-.256	-.482		.183	-.129	-.561	-1.023	
.150	.015	-.165	-.405	-.558		.098	-.137	-.385	-.814	
.250	.005	-.118	-.260	-.365		.031	-.153	-.269	-.535	
.350	-.033	-.133	-.226	-.294		-.039	-.203	-.280	-.366	
.450	-.070	-.186	-.242	-.262		-.102	-.258	-.293	-.276	
.550	-.073	-.168	-.195	-.208		-.122	-.256	-.246	-.235	
.650	-.082	-.156	-.151	-.184		-.131	-.221	-.180	-.227	
.750	-.057	-.097	-.091	-.146		-.094	-.116	-.102	-.200	
.850	-.001	-.001	-.013	-.090		-.015	.006	-.005	-.141	
.900	.034	.040	.031	-.046		.031	.059	.046	-.097	
	<i>Right side</i>									
.025	.054	.022	-.029	.356		-.935	-.017	.407	.624	
.075	-.265	-.113	.054	.327		-.531	-.097	.211	.413	
.150	-.383	-.134	.043	.258		-.357	-.112	.120	.292	
.250	-.261	-.098	.016	.199		-.266	-.132	.054	.201	
.350	-.243	-.134	-.020	.118		-.273	-.171	.003	.131	
.450	-.259	-.179	-.056	.029		-.290	-.217	-.059	.044	
.550	-.206	-.161	-.056	.010		-.265	-.226	-.097	-.008	
.650	-.150	-.136	-.056	-.022		-.182	-.195	-.099	-.039	
.750	-.094	-.087	-.035	-.031		-.087	-.104	-.059	-.051	
.850	-.007	.011	.026	.009		.007	.030	.014	-.023	
.900	.031	.046	.046	.006		.045	.051	.032	-.042	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.447	.735	.454	.057		.794	.923	.586	.221	
.025	.415	-.085	-1.174	-.719		.586	-.274	-.755	-.334	
.075	.258	-.099	-.969	-.739		.301	-.344	-.501	-.338	
.150	.163	-.125	-.235	-.675		.079	-.343	-.408	-.394	
.250	.082	-.167	-.265	-.537		-.001	-.239	-.389	-.378	
.350	-.013	-.249	-.293	-.325						
.450	-.114	-.311	-.313	-.338						
.550	-.197	-.383	-.283	-.362						
.650	-.210	-.285	-.219	-.376		-.387	-.159	-.289	-.354	
.750	-.155	-.094	-.157	-.378		-.157	-.040	-.273	-.345	
.850	-.066	.004	-.094	-.357		-.103	.047	-.235	-.323	
.900	-.037	.051	-.060	-.337		-.085	.081	-.213	-.313	
	<i>Right side</i>									
.025	-1.103	.020	.475	.667		-.541	-.101	.656	.904	
.075	-.802	-.058	.283	.484		-.439	-.264	.339	.588	
.150	-.238	-.070	.197	.370		-.413	-.172	.163	.376	
.250	-.259	-.132	.109	.278						
.350	-.311	-.202	.014	.163						
.450	-.337	-.276	-.079	.052						
.550	-.287	-.315	-.156	-.042						
.650	-.211	-.295	-.178	-.098		-.274	-.156	-.198	-.152	
.750	-.153	-.105	-.133	-.133		-.257	-.051	-.148	-.205	
.850	-.059	.040	-.027	-.104		-.210	.053	-.083	-.197	
.900	-.019	.077	-.004	-.136		-.177	.090	-.083	-.231	

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(o)  $\alpha = 9.7^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	-.337	.980	.292	.448		.179	.749	.559	.590	
.025	-.358	-.050	-.038	.039		-1.279	-.052	.368	-.963	
.075	-.392	-.075	.056	-.162		-1.027	-.099	.199	-.403	
.150	-.484	-.140	.052	-.316		-.725	-.105	.114	-.324	
.250	-.384	-.095	.033	-.220		-.472	-.126	.049	-.262	
.350	-.288	-.117	-.003	-.212		-.341	-.183	-.019	-.271	
.450	-.263	-.171	-.048	-.235		-.279	-.239	-.093	-.298	
.550	-.218	-.165	-.068	-.196		-.252	-.245	-.119	-.254	
.650	-.201	-.159	-.085	-.167		-.245	-.228	-.137	-.194	
.750	-.171	-.096	-.068	-.108		-.230	-.119	-.111	-.117	
.850	-.109	-.002	-.014	-.028		-.171	.010	-.035	-.021	
.900	-.070	.042	.020	.014		-.121	.063	.019	.036	
<i>Right side</i>										
.025	.287	.061	.093	.026		.627	.026	-.859	.422	
.075	.354	-.069	-.183	.086		.419	-.052	-.420	.232	
.150	.305	-.105	-.306	.080		.298	-.074	-.320	.137	
.250	.215	-.071	-.227	.059		.203	-.100	-.245	.068	
.350	.115	-.108	-.233	.014		.120	-.145	-.261	.009	
.450	.027	-.159	-.256	-.037		.030	-.192	-.283	-.059	
.550	.000	-.147	-.211	-.058		-.030	-.214	-.270	-.102	
.650	-.043	-.130	-.164	-.069		-.067	-.200	-.195	-.119	
.750	-.058	-.090	-.102	-.049		-.080	-.132	-.098	-.086	
.850	-.020	.013	-.020	.009		-.060	.031	-.001	-.009	
.900	-.020	.043	.018	.030		-.075	.049	.035	.008	
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.078	.724	.439	.451		.248	.921	.789	.592	
.025	-.745	-.054	.429	-1.090		-.346	-.254	.592	-.727	
.075	-.767	-.070	.276	-.881		-.356	-.317	.315	-.490	
.150	-.693	-.100	.178	-.250		-.412	-.300	.091	-.376	
.250	-.495	-.141	.094	-.254		-.400	-.204	.011	-.362	
.350	-.330	-.222	.003	-.286						
.450	-.360	-.292	-.103	-.319						
.550	-.376	-.364	-.191	-.294						
.650	-.389	-.330	-.216	-.227		-.370	-.159	-.474	-.286	
.750	-.392	-.099	-.185	-.165		-.366	-.039	-.203	-.271	
.850	-.376	.009	-.082	-.104		-.345	.048	-.124	-.237	
.900	-.360	.052	-.047	-.070		-.329	.084	-.111	-.215	
<i>Right side</i>										
.025	.659	.057	-1.052	.482		.884	-.036	-.507	.659	
.075	.479	-.018	-.617	.298		.570	-.213	-.411	.341	
.150	.373	-.037	-.239	.209		.359	-.128	-.388	.176	
.250	.261	-.100	-.245	.115						
.350	.148	-.168	-.296	.020						
.450	.036	-.250	-.335	-.078						
.550	-.064	-.293	-.295	-.165						
.650	-.134	-.308	-.216	-.200		-.259	-.212	-.275	-.349	
.750	-.191	-.147	-.160	-.194		-.237	-.052	-.265	-.185	
.850	-.138	.051	-.073	-.053		-.212	.056	-.231	-.098	
.900	-.164	.084	-.035	-.021		-.252	.095	-.199	-.102	

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(p)  $\alpha = 15.6^\circ$ ;  $M = 0.60$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	-.351	.823	.179	.080	-.500	.273	.746	.333	-.404	-.850
.025	-.612	-.136	.288	.089	-.480	.137	-.093	-.314	-.668	-.893
.075	-.170	-.143	.019	-.084	-.414	.069	-.132	-.291	-.484	-.767
.150	-.038	-.159	-.126	-.184	-.387	.055	-.129	-.259	-.379	-.627
.250	.012	-.116	-.137	-.170	-.326	.041	-.134	-.246	-.325	-.520
.350	.037	-.100	-.162	-.184	-.288	.023	-.161	-.241	-.291	-.437
.450	.037	-.120	-.178	-.184	-.250	.007	-.172	-.226	-.259	-.372
.550	.037	-.102	-.151	-.147	-.200	.016	-.143	-.187	-.218	-.313
.650	.046	-.077	-.108	-.118	-.153	.016	-.111	-.146	-.179	-.265
.750	.048	-.043	-.069	-.077	-.116	.028	-.061	-.087	-.122	-.202
.850	.048	.003	-.022	-.025	-.065	.046	.000	-.024	-.063	-.139
.900	.062	.037	.005	.003	-.028	-.154	-.184	-.144	-.154	-.157
	<i>Right side</i>									
.025	.350	-.100	-.716	-.254	-.098	-.403	-.088	.109	.193	.204
.075	-.004	-.175	-.198	-.068	.033	-.324	-.136	.046	.121	.240
.150	-.145	-.159	-.056	.012	.109	-.285	-.134	.035	.107	.261
.250	-.158	-.118	.014	.062	.163	-.256	-.134	.041	.114	.281
.350	-.181	-.107	.032	.080	.191	-.240	-.145	.030	.105	.272
.450	-.188	-.125	.037	.087	.182	-.228	-.154	.019	.096	.249
.550	-.156	-.102	.037	.089	.159	-.192	-.141	.010	.084	.211
.650	-.111	-.072	.046	.084	.132	-.136	-.107	.021	.080	.173
.750	-.070	-.036	.046	.078	.105	-.083	-.054	.037	.078	.141
.850	-.004	.023	.062	.080	.091	-.013	.016	.057	.080	.109
.900	.014	.050	.064	.080	.071	.014	.025	.048	.064	.080
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	-.009	.675	.060	-.545	-.728	.286	.818	-.178	-.682	-.611
.025	.463	-.068	-.988	-1.129	-.730	.713	-.079	-.954	-.897	-.579
.075	.298	-.091	-.533	-1.020	-.697	.479	-.111	-.836	-.563	-.457
.150	.205	-.104	-.388	-.800	-.656	.254	-.157	-.841	-.545	-.432
.250	.139	-.116	-.318	-.518	-.611	.132	-.154	-.680	-.566	-.441
.350	.089	-.141	-.305	-.382	-.550					
.450	.037	-.161	-.287	-.320	-.489					
.550	.001	-.157	-.241	-.291	-.432					
.650	-.004	-.125	-.171	-.225	-.376	-.054	-.150	-.153	-.329	-.308
.750	.019	-.070	-.097	-.170	-.317	-.029	-.075	-.078	-.254	-.272
.850	.032	-.011	-.026	-.109	-.261	.010	.003	-.013	-.193	-.232
.900	.041	.018	.005	-.075	-.236	.025	.037	.019	-.163	-.218
	<i>Right side</i>									
.025	-1.033	-.038	.471	.621	.615	-.997	-.018	.720	.891	.974
.075	-.582	-.088	.279	.439	.518	-.897	-.107	.446	.630	.784
.150	-.374	-.086	.200	.325	.448	-.859	-.129	.241	.409	.570
.250	-.328	-.109	.139	.253	.385					
.350	-.312	-.136	.080	.175	.326					
.450	-.281	-.147	.041	.130	.263					
.550	-.233	-.141	.014	.089	.213					
.650	-.170	-.111	.010	.066	.157	-.126	-.095	-.011	.032	.060
.750	-.097	-.061	.023	.057	.105	-.074	-.063	-.008	-.020	-.042
.850	.001	.018	.069	.068	.069	.005	.025	.041	-.002	-.080
.900	.032	.041	.069	.057	.028	.030	.057	.057	-.016	-.114

TABLE III. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(a)  $\alpha = 15.8^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.034	.853	.404	.098	-.508	.298	.789	.296	-.226	-.541
.025	-.964	.130	.319	.039	-.463	-.010	-.037	-.304	-.523	-.632
.075	-.568	-.075	-.001	-.125	-.412	-.075	-.126	-.310	-.430	-.576
.150	-.261	-.176	-.190	-.228	-.385	-.056	-.141	-.287	-.370	-.513
.250	-.128	-.129	-.173	-.193	-.327	-.043	-.150	-.276	-.331	-.460
.350	-.061	-.111	-.192	-.193	-.273	-.028	-.177	-.263	-.299	-.404
.450	-.019	-.132	-.207	-.198	-.231	-.034	-.199	-.252	-.267	-.351
.550	.006	-.103	-.158	-.148	-.184	-.011	-.161	-.213	-.231	-.302
.650	.024	-.067	-.116	-.114	-.145	.000	-.128	-.164	-.195	-.258
.750	.034	-.032	-.083	-.083	-.110	.016	-.067	-.104	-.145	-.207
.850	.046	.007	-.034	-.039	-.066	.042	.009	-.036	-.084	-.151
.900	.052	.043	-.002	-.008	-.033	-.143	-.167	-.133	-.136	-.137
	<i>Right side</i>									
.025	.384	-.801	-.939	-.417	-.309	-.401	-.268	-.054	.007	.069
.075	-.025	-.400	-.576	-.240	-.170	-.351	-.253	-.092	-.030	.115
.150	-.202	-.236	-.280	-.151	-.045	-.317	-.188	-.073	-.013	.172
.250	-.199	-.129	-.142	-.073	.072	-.277	-.140	-.036	.022	.231
.350	-.214	-.075	-.063	-.011	.142	-.265	-.121	-.019	.039	.253
.450	-.214	-.061	-.020	.031	.177	-.250	-.126	-.017	.052	.250
.550	-.167	-.032	.007	.051	.169	-.218	-.109	-.007	.063	.231
.650	-.117	-.014	.024	.066	.148	-.161	-.070	.010	.072	.204
.750	-.076	.009	.036	.072	.124	-.096	-.020	.031	.077	.164
.850	-.016	.048	.057	.081	.097	-.022	.043	.058	.084	.130
.900	.001	.060	.060	.075	.080	.003	.052	.051	.067	.086
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.366	.720	.461	.066	-.397	.600	.877	.399	-.140	-.656
.025	.394	-.035	-1.007	-.946	-.645	.656	-.090	-.714	-.848	-.582
.075	.236	-.078	-.504	-.822	-.621	.414	-.167	-.533	-.420	-.434
.150	.155	-.103	-.358	-.657	-.591	.192	-.203	-.483	-.376	-.409
.250	.104	-.120	-.289	-.461	-.553	.072	-.191	-.501	-.389	-.412
.350	.048	-.161	-.289	-.343	-.513					
.450	-.005	-.197	-.301	-.286	-.468					
.550	-.041	-.196	-.273	-.269	-.413					
.650	-.034	-.152	-.201	-.249	-.363	-.117	-.161	-.261	-.358	-.276
.750	-.010	-.073	-.120	-.208	-.312	-.073	-.058	-.208	-.326	-.249
.850	.018	-.003	-.046	-.161	-.271	-.037	.037	-.143	-.276	-.219
.900	.027	.037	-.007	-.131	-.252	-.023	.075	-.107	-.255	-.208
	<i>Right side</i>									
.025	-1.093	-.059	.413	.614	.710	-.730	.016	.669	.889	1.042
.075	-.522	-.093	.225	.401	.579	-.533	-.128	.380	.614	.837
.150	-.338	-.064	.163	.308	.507	-.519	-.132	.186	.383	.614
.250	-.298	-.084	.108	.236	.444					
.350	-.309	-.117	.045	.163	.374					
.450	-.309	-.137	.004	.110	.305					
.550	-.271	-.134	-.025	.074	.247					
.650	-.203	-.097	-.025	.054	.184	-.233	-.093	-.051	-.016	.069
.750	-.120	-.035	-.001	.039	.122	-.197	-.035	-.057	-.072	-.039
.850	-.025	.051	.049	.049	.077	-.128	.063	-.001	-.057	-.098
.900	.016	.077	.063	.039	.027	-.096	.093	.007	-.081	-.157

TABLE III.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Concluded

(r)  $\alpha = 15.9^\circ$ ;  $M = 0.85$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.410	.905	.585	.091	-.584	.403	.783	.322	-.237	-.517
.025	-1.053	.008	.238	-.095	-.547	.207	-.150	-.425	-.597	-.641
.075	-.489	-.200	-.054	-.225	-.496	.035	-.224	-.409	-.501	-.587
.150	-.227	-.287	-.279	-.323	-.452	-.024	-.209	-.364	-.432	-.531
.250	-.155	-.202	-.231	-.262	-.382	-.061	-.198	-.318	-.377	-.482
.350	-.125	-.152	-.230	-.241	-.315	-.084	-.214	-.295	-.336	-.428
.450	-.102	-.157	-.235	-.231	-.256	-.102	-.231	-.279	-.303	-.375
.550	-.067	-.119	-.176	-.170	-.200	-.081	-.194	-.240	-.258	-.327
.650	-.043	-.084	-.131	-.129	-.156	-.067	-.152	-.192	-.221	-.281
.750	-.013	-.044	-.097	-.105	-.122	-.038	-.089	-.129	-.170	-.225
.850	.018	.005	-.043	-.058	-.075	.010	.007	-.054	-.110	-.176
.900	.045	.038	-.012	-.028	-.040	-.156	-.166	-.139	-.145	-.141
<i>Right side</i>										
.025	.271	-.704	-1.064	-.478	-.412	-.940	-.270	-.074	-.044	.001
.075	-.159	-.851	-.656	-.313	-.270	-.700	-.306	-.153	-.096	.034
.150	-.436	-.317	-.340	-.238	-.134	-.475	-.253	-.144	-.086	.098
.250	-.298	-.225	-.220	-.163	-.011	-.288	-.200	-.111	-.045	.163
.350	-.246	-.149	-.151	-.089	.072	-.270	-.176	-.088	-.023	.186
.450	-.223	-.120	-.091	-.034	.126	-.264	-.178	-.080	-.006	.203
.550	-.165	-.070	-.043	-.001	.135	-.227	-.152	-.057	.013	.193
.650	-.122	-.040	-.012	.030	.129	-.172	-.102	-.025	.036	.175
.750	-.071	-.009	.011	.043	.112	-.100	-.041	.002	.050	.148
.850	-.007	.036	.042	.058	.091	-.016	.031	.039	.061	.116
.900	.014	.050	.046	.054	.072	.011	.048	.035	.043	.072
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.337	.721	.506	.172	-.261	.631	.892	.507	-.001	-.615
.025	.408	-.088	-1.169	-.966	-.651	.651	-.109	-.748	-.906	-.615
.075	.241	-.118	-.521	-.830	-.624	.401	-.217	-.484	-.410	-.449
.150	.150	-.133	-.360	-.666	-.601	.161	-.248	-.418	-.362	-.412
.250	.085	-.150	-.292	-.489	-.563	.04	-.226	-.428	-.367	-.423
.350	.014	-.195	-.291	-.360	-.521					
.450	-.063	-.242	-.312	-.298	-.475					
.550	-.109	-.246	-.286	-.272	-.426					
.650	-.105	-.180	-.221	-.262	-.382	-.191	-.177	-.301	-.367	-.304
.750	-.073	-.091	-.151	-.244	-.338	-.143	-.064	-.268	-.349	-.276
.850	-.033	-.012	-.078	-.207	-.296	-.097	.039	-.214	-.318	-.244
.900	-.017	.028	-.039	-.180	-.279	-.087	.073	-.180	-.299	-.232
<i>Right side</i>										
.025	-1.335	-.077	.378	.596	.727	-.813	-.007	.647	.882	1.054
.075	-.828	-.123	.187	.382	.578	-.453	-.188	.343	.596	.837
.150	-.274	-.102	.129	.284	.494	-.443	-.188	.139	.353	.605
.250	-.271	-.125	.069	.209	.426					
.350	-.298	-.167	-.005	.128	.356					
.450	-.307	-.187	-.049	.068	.288					
.550	-.280	-.184	-.084	.027	.226					
.650	-.222	-.140	-.077	.010	.172	-.279	-.120	-.107	-.065	.062
.750	-.158	-.067	-.046	-.003	.114	-.259	-.051	-.111	-.122	-.041
.850	-.063	.039	.019	.012	.067	-.205	.056	-.046	-.112	-.105
.900	-.019	.073	.035	-.004	.011	-.168	.091	-.043	-.145	-.173



TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$

(a)  $\alpha = -9.4^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.724	.836	.562	-.168	-1.028	.460	.824	.548	-.092	-.687
.025	.246	-.112	-.546	-1.005	-2.178	.330	-.094	-.594	-1.159	-1.069
.075	.100	-.130	-.363	-.622	-1.082	.119	-.153	-.450	-.929	-1.053
.150	-.014	-.180	-.336	-.475	-.674	.009	-.185	-.363	-.565	-.954
.250	-.052	-.158	-.244	-.322	-.404	-.082	-.222	-.343	-.420	-.817
.350	-.086	-.164	-.228	-.255	-.317	-.150	-.260	-.359	-.374	-.664
.450	-.130	-.190	-.233	-.241	-.281	-.200	-.299	-.359	-.342	-.543
.550	-.141	-.180	-.201	-.193	-.242	-.200	-.263	-.306	-.294	-.461
.650	-.109	-.142	-.149	-.152	-.212	-.173	-.217	-.244	-.251	-.422
.750	-.077	-.098	-.091	-.104	-.190	-.123	-.153	-.167	-.189	-.381
.850	-.025	-.034	-.032	-.044	-.135	-.050	-.066	-.073	-.118	-.317
.900	.005	.007	.007	-.005	-.091	.000	-.023	-.025	-.072	-.270
	<i>Right side</i>									
.025	-.478	-.078	.283	.541	.776	-.711	-.114	.340	.591	.760
.075	-.414	-.171	.078	.275	.479	-.474	-.183	.109	.311	.491
.150	-.348	-.187	-.012	.132	.304	-.389	-.199	.002	.153	.313
.250	-.269	-.174	-.046	.059	.185	-.346	-.231	-.087	.032	.167
.350	-.246	-.185	-.096	-.026	.082	-.353	-.265	-.149	-.065	.057
.450	-.250	-.208	-.137	-.085	-.007	-.351	-.285	-.196	-.129	-.037
.550	-.209	-.180	-.133	-.104	-.043	-.314	-.267	-.194	-.157	-.091
.650	-.150	-.139	-.107	-.088	-.053	-.239	-.212	-.167	-.147	-.103
.750	-.096	-.089	-.071	-.067	-.046	-.159	-.135	-.105	-.101	-.085
.850	-.011	-.009	.000	-.005	-.009	-.055	-.041	-.012	-.035	-.055
.900	.018	.027	.020	.004	.000	-.014	-.023	-.012	-.030	-.075
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.422	.838	.530	.022	-.340	.358	.607	.114	.057	-.098
.025	.340	-.171	-.859	-1.106	-.817	.073	-.909	-1.596	-.551	-.505
.075	.105	-.240	-.598	-1.010	-.852	-.278	-.747	-1.603	-.501	-.496
.150	-.061	-.306	-.550	-.746	-.804	-.417	-.699	-1.295	-.484	-.498
.250	-.178	-.368	-.537	-.526	-.649	-.380	-.557	-.747	-.480	-.507
.350	-.264	-.413	-.537	-.443	-.580					
.450	-.317	-.434	-.521	-.411	-.564					
.550	-.310	-.400	-.461	-.395	-.573					
.650	-.255	-.308	-.359	-.381	-.580	-.307	-.356	-.388	-.507	-.518
.750	-.171	-.217	-.251	-.386	-.573	-.187	-.215	-.254	-.496	-.491
.850	-.105	-.126	-.149	-.368	-.523	-.093	-.098	-.128	-.459	-.448
.900	-.073	-.087	-.101	-.342	-.505	-.055	-.048	-.075	-.434	-.425
	<i>Right side</i>									
.025	-.854	-.176	.354	.598	.751	-1.524	-.861	.082	.360	.609
.075	-.642	-.270	.078	.298	.479	-1.556	-.760	-.295	-.062	.185
.150	-.533	-.308	-.062	.114	.288	-1.312	-.685	-.425	-.237	-.028
.250	-.551	-.384	-.187	-.046	.116					
.350	-.558	-.427	-.274	-.163	-.023					
.450	-.519	-.432	-.317	-.241	-.130					
.550	-.455	-.400	-.304	-.264	-.183					
.650	-.358	-.297	-.247	-.246	-.194	-.332	-.263	-.199	-.230	-.196
.750	-.253	-.210	-.158	-.214	-.185	-.241	-.199	-.162	-.253	-.244
.850	-.125	-.094	-.066	-.124	-.158	-.107	-.075	-.064	-.195	-.206
.900	-.059	-.041	-.030	-.134	-.187	-.050	-.023	-.016	-.205	-.219

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -8^\circ$  - Continued

(b)  $\alpha = -9.6^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.847	.883	.714	.372	-.100	.664	.866	.729	.381	-.089
.025	.240	-.102	-.554	-.863	-1.374	.324	-.073	-.942	-1.568	-.904
.075	.104	-.130	-.382	-.784	-1.103	.122	-.148	-.452	-1.345	-.899
.150	-.018	-.192	-.362	-.519	-.851	.014	-.184	-.373	-.398	-.816
.250	-.055	-.167	-.254	-.298	-.606	-.090	-.225	-.350	-.314	-.778
.350	-.102	-.184	-.244	-.238	-.429	-.189	-.314	-.387	-.323	-.622
.450	-.177	-.231	-.266	-.232	-.345	-.284	-.384	-.415	-.326	-.473
.550	-.205	-.233	-.242	-.206	-.273	-.313	-.355	-.362	-.288	-.371
.650	-.186	-.193	-.190	-.162	-.235	-.283	-.292	-.291	-.258	-.320
.750	-.137	-.126	-.122	-.123	-.204	-.197	-.190	-.201	-.212	-.307
.850	-.061	-.046	-.037	-.062	-.151	-.087	-.081	-.093	-.144	-.300
.900	-.016	-.002	.005	-.022	-.118	-.030	-.028	-.033	-.105	-.271
	<i>Right side</i>									
.025	-.424	-.034	.295	.537	.767	-.924	-.081	.349	.580	.758
.075	-.417	-.154	.090	.281	.483	-.456	-.164	.119	.313	.500
.150	-.363	-.187	-.011	.136	.306	-.384	-.199	.005	.157	.321
.250	-.266	-.173	-.053	.054	.184	-.348	-.234	-.090	.028	.167
.350	-.265	-.198	-.116	-.037	.066	-.378	-.296	-.184	-.084	.040
.450	-.278	-.239	-.183	-.126	-.046	-.402	-.357	-.272	-.193	-.097
.550	-.239	-.231	-.195	-.161	-.109	-.367	-.349	-.292	-.250	-.180
.650	-.179	-.181	-.172	-.162	-.130	-.287	-.281	-.265	-.255	-.213
.750	-.116	-.123	-.128	-.135	-.135	-.183	-.167	-.166	-.191	-.191
.850	-.024	-.025	-.029	-.058	-.091	-.069	-.054	-.053	-.103	-.145
.900	.014	.011	-.001	-.032	-.074	-.025	-.028	-.035	-.088	-.166
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.677	.901	.749	.417	.025	.658	.739	.603	.452	.184
.025	.348	-.110	-1.093	-1.600	-.899	-.009	-.854	-1.414	-.448	-.426
.075	.121	-.196	-.478	-1.321	-.919	-.333	-.930	-1.315	-.413	-.421
.150	-.042	-.283	-.510	-.633	-.848	-.465	-.848	-1.265	-.412	-.427
.250	-.200	-.395	-.530	-.401	-.582	-.405	-.586	-.851	-.403	-.430
.350	-.345	-.526	-.600	-.327	-.430					
.450	-.465	-.545	-.560	-.344	-.453					
.550	-.519	-.549	-.534	-.370	-.477					
.650	-.427	-.367	-.429	-.388	-.505	-.397	-.408	-.542	-.423	-.462
.750	-.242	-.267	-.318	-.416	-.528	-.268	-.263	-.414	-.430	-.456
.850	-.147	-.155	-.195	-.426	-.521	-.159	-.149	-.279	-.416	-.433
.900	-.107	-.110	-.137	-.429	-.524	-.122	-.107	-.213	-.407	-.427
	<i>Right side</i>									
.025	-1.083	-.102	.371	.612	.777	-1.345	-.605	.021	.360	.636
.075	-.491	-.220	.102	.320	.506	-1.296	-.873	-.379	-.070	.202
.150	-.482	-.278	-.043	.142	.319	-1.232	-.783	-.516	-.277	-.039
.250	-.534	-.401	-.208	-.050	.123					
.350	-.635	-.539	-.345	-.215	-.062					
.450	-.554	-.517	-.444	-.353	-.221					
.550	-.524	-.525	-.470	-.435	-.327					
.650	-.435	-.431	-.428	-.429	-.359	-.432	-.349	-.292	-.336	-.304
.750	-.305	-.263	-.227	-.292	-.301	-.403	-.272	-.231	-.302	-.301
.850	-.158	-.114	-.105	-.190	-.230	-.254	-.131	-.120	-.223	-.230
.900	-.099	-.067	-.065	-.196	-.251	-.185	-.076	-.084	-.226	-.242

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(c)  $\alpha = -9.7^\circ$ ;  $M = 0.85$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.881	.906	.760	.478	.053	.707	.888	.772	.498	.045
.025	.245	-.104	-.539	-.827	-1.312	.328	-.077	-.987	-1.088	-1.080
.075	.110	-.134	-.420	-.784	-1.035	.129	-.161	-.397	-1.217	-1.026
.150	-.016	-.205	-.387	-.662	-.815	.014	-.189	-.373	-1.074	-.726
.250	-.053	-.178	-.259	-.354	-.633	-.085	-.236	-.342	-.337	-.735
.350	-.106	-.193	-.247	-.262	-.442	-.192	-.308	-.388	-.309	-.670
.450	-.187	-.254	-.285	-.249	-.356	-.313	-.421	-.447	-.326	-.528
.550	-.227	-.280	-.266	-.223	-.302	-.366	-.456	-.418	-.346	-.412
.650	-.225	-.249	-.208	-.186	-.262	-.359	-.388	-.309	-.309	-.349
.750	-.171	-.169	-.135	-.132	-.223	-.248	-.225	-.214	-.214	-.308
.850	-.078	-.070	-.044	-.058	-.165	-.111	-.101	-.108	-.113	-.285
.900	-.032	-.020	-.001	-.018	-.129	-.046	-.034	-.042	-.055	-.265
	<i>Right side</i>									
.025	-.412	-.021	.305	.541	.772	-.953	-.074	.356	.579	.760
.075	-.433	-.158	.094	.278	.487	-.470	-.163	.126	.314	.504
.150	-.387	-.196	-.006	.138	.308	-.384	-.192	.013	.162	.325
.250	-.277	-.181	-.054	.055	.192	-.347	-.232	-.088	.037	.179
.350	-.273	-.215	-.119	-.035	.069	-.383	-.308	-.188	-.084	.047
.450	-.293	-.269	-.200	-.137	-.053	-.440	-.394	-.301	-.206	-.100
.550	-.263	-.283	-.235	-.184	-.127	-.416	-.442	-.366	-.288	-.203
.650	-.198	-.233	-.224	-.197	-.168	-.304	-.365	-.359	-.319	-.271
.750	-.129	-.162	-.169	-.176	-.192	-.202	-.192	-.237	-.242	-.279
.850	-.035	-.048	-.055	-.072	-.138	-.081	-.074	-.068	-.102	-.209
.900	.005	-.010	-.021	-.037	-.118	-.045	-.047	-.052	-.068	-.204
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.716	.923	.794	.519	.138	.697	.780	.677	.455	.223
.025	.356	-.095	-1.068	-1.427	-.798	.014	-.725	-1.316	-.994	-.419
.075	.137	-.183	-.781	-1.274	-.815	-.324	-.859	-1.373	-.933	-.408
.150	-.035	-.267	-.468	-1.160	-.800	-.434	-.803	-1.140	-.949	-.409
.250	-.191	-.369	-.502	-.852	-.764	-.459	-.681	-.784	-.966	-.417
.350	-.346	-.514	-.587	-.278	-.548					
.450	-.499	-.638	-.672	-.492	-.436					
.550	-.578	-.668	-.502	-.579	-.439					
.650	-.585	-.627	-.448	-.404	-.462	-.598	-.425	-.574	-.474	-.448
.750	-.300	-.205	-.359	-.369	-.484	-.298	-.340	-.474	-.519	-.448
.850	-.152	-.178	-.227	-.292	-.494	-.199	-.237	-.330	-.484	-.438
.900	-.131	-.138	-.169	-.244	-.504	-.167	-.183	-.310	-.449	-.432
	<i>Right side</i>									
.025	-1.061	-.075	.286	.617	.781	-1.270	-.519	.047	.403	.670
.075	-.740	-.195	.121	.332	.521	-1.250	-.824	-.357	-.042	.226
.150	-.452	-.250	-.021	.153	.333	-1.198	-.719	-.471	-.251	-.016
.250	-.523	-.379	-.187	-.038	.139					
.350	-.628	-.526	-.354	-.207	-.049					
.450	-.654	-.655	-.503	-.370	-.211					
.550	-.490	-.636	-.584	-.468	-.349					
.650	-.475	-.534	-.549	-.525	-.458	-.463	-.455	-.566	-.689	-.736
.750	-.341	-.286	-.401	-.503	-.531	-.467	-.345	-.315	-.322	-.525
.850	-.185	-.139	-.106	-.108	-.279	-.340	-.209	-.166	-.176	-.219
.900	-.125	-.100	-.081	-.101	-.252	-.271	-.155	-.123	-.162	-.213

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(d)  $\alpha = -9.8^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.913	.931	.807	.556	.126	.752	.902	.807	.575	.136
.025	.257	-.081	-.473	-.803	-1.143	.339	-.049	-.887	-1.188	-.691
.075	.122	-.114	-.455	-.732	-.946	.143	-.142	-.889	-1.092	-.691
.150	-.005	-.195	-.415	-.672	-.782	.035	-.170	-.420	-1.004	-.705
.250	-.042	-.163	-.270	-.616	-.602	-.062	.203	-.290	-.822	-.702
.350	-.092	-.175	-.233	-.325	-.539	-.171	.281	-.360	-.444	-.653
.450	-.182	-.245	-.273	-.254	-.512	-.300	.384	-.425	-.215	-.587
.550	-.260	-.308	-.302	-.182	-.433	-.376	.467	-.473	-.211	-.522
.650	-.291	-.323	-.271	-.167	-.354	-.457	.523	-.456	-.283	-.446
.750	-.332	-.304	-.171	-.140	-.284	-.486	.508	-.246	-.259	-.356
.850	-.178	-.096	-.057	-.080	-.221	-.297	.100	-.104	-.143	-.325
.900	-.078	-.025	-.009	-.039	-.196	-.107	.013	-.040	-.080	-.304
	<i>Right side</i>									
.025	-.366	-.003	.317	.554	.800	-.869	-.057	.356	.580	.772
.075	-.467	-.140	.101	.294	.515	-.687	.154	.141	.322	.522
.150	-.438	-.193	-.001	.150	.335	-.483	.175	.028	.181	.351
.250	-.291	-.167	-.042	.071	.213	-.293	.209	-.068	.050	.198
.350	-.257	-.195	-.106	-.024	.089	-.359	.284	-.171	-.065	.070
.450	-.289	-.254	-.193	-.128	-.033	-.435	.375	-.288	-.199	-.080
.550	-.303	-.312	-.261	-.199	-.120	-.485	.467	-.376	-.298	-.191
.650	-.254	-.303	-.282	-.244	-.181	-.432	.517	-.445	-.377	-.273
.750	-.165	-.296	-.351	-.329	-.272	-.228	.472	-.476	-.436	-.349
.850	-.051	-.068	-.158	-.210	-.257	-.094	.037	-.331	-.359	-.357
.900	-.008	-.020	-.070	-.132	-.316	-.046	.012	-.081	-.250	-.430
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.756	.938	.838	.587	.197	.748	.822	.731	.531	.254
.025	.377	-.056	-.965	-1.252	-.782	.075	.597	-1.126	-.894	-.436
.075	.169	-.146	-.808	-1.124	-.795	-.262	.753	-1.322	-.838	-.426
.150	.004	-.229	-.666	-1.057	-.819	-.396	.747	-1.169	-.835	-.426
.250	-.149	-.327	-.386	-1.009	-.924	-.434	.734	-.839	-.846	-.432
.350	-.307	-.460	-.519	-.746	-.569					
.450	-.462	-.602	-.614	-.307	-.445					
.550	-.601	-.703	-.628	-.449	-.450					
.650	-.659	-.738	-.600	-.485	-.468	-.847	.717	-.617	-.350	-.450
.750	-.704	-.711	-.427	-.387	-.485	-.774	.469	-.524	-.504	-.453
.850	-.610	-.092	-.209	-.339	-.490	-.451	.308	-.461	-.532	-.449
.900	-.260	-.033	-.177	-.306	-.505	-.340	.261	-.428	-.514	-.446
	<i>Right side</i>									
.025	-.952	-.045	.404	.623	.795	-1.181	.459	.101	.439	.716
.075	-.806	-.167	.150	.348	.539	-1.226	.749	-.312	-.005	.282
.150	-.632	-.218	.011	.174	.363	-1.115	.666	-.392	-.222	.037
.250	-.403	-.334	-.153	-.002	.170					
.350	-.561	-.472	-.314	-.193	-.012					
.450	-.636	-.621	-.459	-.335	-.173					
.550	-.631	-.695	-.586	-.488	-.310					
.650	-.584	-.717	-.674	-.553	-.430	-.515	.667	-.783	-.797	-.693
.750	-.424	-.663	-.702	-.649	-.550	-.533	.480	-.707	-.850	-.746
.850	-.187	-.046	-.617	-.648	-.557	-.436	.281	-.404	-.764	-.738
.900	-.142	-.012	-.197	-.611	-.583	-.392	.231	-.304	-.579	-.738

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(e)  $\alpha = -9.8^\circ$ ;  $M = 0.92$

$\frac{x}{c_v}$	<i>C<sub>p</sub> for -</i>					<i>C<sub>p</sub> for -</i>				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.924	.942	.815	.555	.182	.753	.908	.814	.569	.172
.025	.283	-.082	-.476	-.859	-1.189	.355	-.053	-.885	-1.178	-.912
.075	.143	-.116	-.489	-.746	-.941	.159	-.149	-.737	-1.088	-.891
.150	.011	-.199	-.455	-.700	-.839	.045	-.174	-.509	-1.019	-.777
.250	-.027	-.168	-.436	-.692	-.659	-.049	-.207	-.469	-.932	-.735
.350	-.082	-.182	-.246	-.420	-.519	-.156	-.290	-.336	-.701	-.658
.450	-.173	-.247	-.277	-.315	-.450	-.281	-.380	-.411	-.322	-.560
.550	-.258	-.314	-.318	-.187	-.380	-.367	-.473	-.473	-.149	-.486
.650	-.291	-.342	-.300	-.150	-.342	-.447	-.548	-.463	-.219	-.434
.750	-.355	-.390	-.215	-.142	-.312	-.486	-.566	-.331	-.260	-.392
.850	-.287	-.199	-.104	-.098	-.271	-.480	-.321	-.195	-.188	-.396
.900	-.145	-.062	-.045	-.061	-.244	-.284	-.063	-.086	-.131	-.374
	<i>Right side</i>									
.025	-.370	.000	.323	.573	.794	-.865	-.057	.366	.595	.770
.075	-.502	-.149	.109	.315	.514	-.736	-.159	.137	.340	.523
.150	-.460	-.199	-.001	.167	.335	-.557	-.178	.030	.194	.352
.250	-.383	-.172	-.041	.083	.213	-.336	-.207	-.061	.064	.202
.350	-.258	-.196	-.111	-.015	.086	-.329	-.288	-.169	-.051	.073
.450	-.280	-.260	-.196	-.121	-.036	-.410	-.368	-.287	-.192	-.075
.550	-.304	-.329	-.274	-.199	-.126	-.480	-.472	-.390	-.285	-.185
.650	-.263	-.319	-.290	-.240	-.182	-.391	-.533	-.447	-.368	-.273
.750	-.186	-.390	-.385	-.343	-.283	-.297	-.536	-.485	-.424	-.342
.850	-.070	-.161	-.315	-.303	-.266	-.128	-.271	-.520	-.426	-.367
.900	-.026	-.054	-.232	-.277	-.349	-.066	-.023	-.468	-.449	-.435
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.758	.946	.843	.589	.224	.761	.833	.744	.541	.272
.025	.397	-.052	-.959	-1.232	-.868	.110	-.541	-1.129	-.878	-.467
.075	.186	-.141	-.814	-1.109	-.870	-.227	-.730	-1.332	-.833	-.455
.150	.023	-.222	-.760	-1.053	-.839	-.367	-.728	-1.216	-.830	-.454
.250	-.123	-.319	-.634	-1.028	-.714	-.410	-.738	-.850	-.834	-.460
.350	-.281	-.448	-.451	-.957	-.503					
.450	-.437	-.591	-.567	-.449	-.481					
.550	-.583	-.709	-.506	-.394	-.490					
.650	-.650	-.763	-.588	-.463	-.506	-.873	-.783	-.644	-.328	-.482
.750	-.717	-.763	-.626	-.403	-.524	-.918	-.577	-.584	-.509	-.484
.850	-.684	-.541	-.364	-.378	-.527	-.661	-.418	-.544	-.561	-.477
.900	-.662	-.138	-.221	-.377	-.535	-.485	-.346	-.510	-.555	-.476
	<i>Right side</i>									
.025	-.948	-.041	.418	.635	.797	-1.119	-.393	.130	.466	.725
.075	-.814	-.167	.158	.362	.541	-1.141	-.709	-.293	.021	.284
.150	-.719	-.213	.024	.195	.364	-1.071	-.667	-.376	-.200	.042
.250	-.587	-.326	-.140	.013	.178					
.350	-.476	-.462	-.303	-.162	-.004					
.450	-.574	-.613	-.449	-.315	-.171					
.550	-.514	-.701	-.588	-.463	-.304					
.650	-.599	-.751	-.683	-.535	-.418	-.516	-.701	-.871	-.787	-.688
.750	-.558	-.730	-.721	-.636	-.541	-.575	-.598	-.910	-.853	-.737
.850	-.238	-.436	-.710	-.647	-.555	-.485	-.380	-.689	-.826	-.728
.900	-.154	-.049	-.681	-.665	-.587	-.451	-.309	-.497	-.808	-.734

TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(f)  $\alpha = 0^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.611	.605	.664	-.455	-1.262	.287	.286	.751	-.386	-.987
.025	.277	.275	-.103	-1.117	-1.934	.344	.339	-.091	-1.057	-.987
.075	.130	.132	-.117	-.675	-1.470	.151	.153	-.139	-.927	-.971
.150	.031	.036	-.144	-.474	-.812	.048	.050	-.158	-.682	-.890
.250	.015	.013	-.110	-.300	-.422	-.024	-.026	-.171	-.420	-.823
.350	-.033	-.042	-.128	-.240	-.345	-.088	-.085	-.206	-.344	-.715
.450	-.079	-.079	-.151	-.217	-.306	-.127	-.127	-.222	-.302	-.595
.550	-.081	-.079	-.133	-.170	-.308	-.132	-.129	-.194	-.258	-.486
.650	-.065	-.062	-.094	-.131	-.251	-.113	-.108	-.155	-.221	-.422
.750	-.038	-.037	-.043	-.089	-.172	-.065	-.069	-.094	-.166	-.350
.850	.008	.006	.016	-.029	-.096	.002	-.001	-.009	-.092	-.271
.900	.031	.034	.046	-.001	-.050	.027	.029	.030	-.041	-.214
	<i>Right side</i>									
.025	-.481	-.480	-.028	.610	.821	-.661	-.659	-.069	.607	.731
.075	-.403	-.407	-.119	.350	.542	-.465	-.466	-.128	.362	.523
.150	-.327	-.317	-.133	.214	.371	-.369	-.368	-.149	.216	.375
.250	-.238	-.239	-.117	.128	.257	-.318	-.324	-.167	.110	.239
.350	-.221	-.214	-.135	.054	.158	-.316	-.306	-.192	.033	.154
.450	-.212	-.207	-.153	-.013	.080	-.258	-.260	-.180	-.048	.038
.550	-.159	-.161	-.126	-.025	.054	-.263	-.260	-.196	-.071	.017
.650	-.111	-.113	-.091	-.015	.040	-.192	-.186	-.146	-.059	-.001
.750	-.061	-.049	-.041	.001	.036	-.109	-.101	-.078	-.034	-.001
.850	.020	.018	.025	.038	.057	-.012	-.007	.018	.017	.010
.900	.048	.041	.052	.050	.052	.022	.027	.036	.017	-.013
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.185	.188	.742	-.305	-.650	.358	.364	.660	.045	-.382
.025	.374	.373	-.142	-.974	-.763	.266	.272	-.738	-.490	-.447
.075	.162	.165	-.183	-.904	-.805	-.038	-.033	-.528	-.457	-.454
.150	.038	.041	-.226	-.738	-.821	-.212	-.202	-.507	-.444	-.454
.250	-.065	-.060	-.265	-.481	-.740	-.240	-.232	-.457	-.448	-.472
.350	-.148	-.138	-.299	-.388	-.555					
.450	-.192	-.189	-.313	-.365	-.519					
.550	-.194	-.193	-.285	-.360	-.530					
.650	-.159	-.159	-.219	-.358	-.539	-.185	-.177	-.224	-.439	-.491
.750	-.109	-.111	-.119	-.351	-.525	-.107	-.099	-.110	-.432	-.456
.850	-.028	-.028	-.048	-.321	-.468	-.026	-.033	-.021	-.388	-.410
.900	-.008	-.010	-.014	-.295	-.447	.008	.013	.025	-.363	-.389
	<i>Right side</i>									
.025	-.900	-.881	-.089	.617	.719	-1.394	-1.395	-.496	.580	.816
.075	-.587	-.583	-.164	.381	.528	-1.375	-1.377	-.505	.172	.408
.150	-.472	-.468	-.187	.228	.382	-1.152	-1.143	-.445	.008	.204
.250	-.470	-.466	-.256	.087	.239					
.350	-.463	-.468	-.288	-.018	.114					
.450	-.421	-.416	-.292	-.094	.022					
.550	-.355	-.351	-.270	-.131	-.043					
.650	-.261	-.257	-.203	-.129	-.073	-.182	-.175	-.185	-.161	-.144
.750	-.157	-.150	-.123	-.108	-.089	-.123	-.122	-.105	-.182	-.172
.850	-.040	-.037	-.018	-.069	-.098	-.021	-.017	.000	-.143	-.156
.900	.006	.009	.018	-.078	-.128	.025	.027	.050	-.157	-.179

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(g)  $\alpha = 0^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.717	.677	.545	.202	-.275	.498	.780	.561	.088	-.361
.025	.279	-.076	-.521	-1.111	-1.021	.339	-.084	-.900	-1.241	-.663
.075	.146	-.101	-.365	-.828	-.886	.157	-.137	-.431	-1.048	-.670
.150	.042	-.148	-.323	-.463	-.790	.057	-.157	-.333	-.811	-.687
.250	.016	-.116	-.219	-.281	-.637	-.027	-.186	-.291	-.480	-.687
.350	-.047	-.146	-.200	-.228	-.531	-.104	-.241	-.303	-.299	-.624
.450	-.102	-.185	-.202	-.203	-.471	-.174	-.284	-.303	-.240	-.531
.550	-.108	-.165	-.158	-.162	-.382	-.183	-.263	-.252	-.217	-.445
.650	-.101	-.128	-.110	-.127	-.295	-.166	-.208	-.190	-.217	-.367
.750	-.065	-.071	-.051	-.091	-.223	-.108	-.111	-.112	-.188	-.303
.850	-.004	.005	.018	-.038	-.140	-.018	-.007	-.017	-.121	-.254
.900	.028	.046	.054	-.001	-.091	.025	.042	.032	-.071	-.226
	<i>Right side</i>									
.025	-.416	.002	.331	.599	.818	-.957	-.042	.372	.605	.745
.075	-.393	-.101	.138	.356	.550	-.428	-.122	.161	.371	.535
.150	-.320	-.131	.051	.219	.385	-.347	-.145	.062	.233	.380
.250	-.231	-.117	.012	.143	.275	-.292	-.174	-.019	.122	.261
.350	-.217	-.150	-.051	.058	.169	-.300	-.220	-.091	.032	.160
.450	-.209	-.182	-.107	-.019	.072	-.269	-.231	-.141	-.056	.045
.550	-.166	-.159	-.110	-.044	.034	-.266	-.255	-.181	-.098	-.009
.650	-.105	-.117	-.091	-.048	.008	-.185	-.189	-.153	-.106	-.038
.750	-.042	-.061	-.057	-.039	-.006	-.097	-.091	-.086	-.077	-.047
.850	.028	.020	.015	.011	.011	.004	.013	.000	-.024	-.047
.900	.063	.054	.036	.026	.005	.037	.046	.022	-.019	-.073
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.461	.777	.514	.161	-.364	.648	.742	.555	.403	-.107
.025	.370	-.111	-1.040	-.694	-.765	.175	-.750	-.576	-.374	-.381
.075	.178	-.165	-.521	-.702	-.773	-.091	-.598	-.529	-.354	-.385
.150	.042	-.222	-.411	-.656	-.776	-.286	-.592	-.492	-.352	-.387
.250	-.079	-.295	-.368	-.457	-.765	-.300	-.535	-.474	-.355	-.395
.350	-.189	-.385	-.378	-.330	-.509					
.450	-.281	-.436	-.376	-.340	-.436					
.550	-.304	-.402	-.332	-.367	-.437					
.650	-.249	-.258	-.257	-.389	-.459	-.272	-.208	-.335	-.387	-.434
.750	-.148	-.134	-.188	-.395	-.474	-.165	-.081	-.327	-.378	-.428
.850	-.059	-.029	-.123	-.371	-.468	-.099	.017	-.288	-.357	-.405
.900	-.039	.013	-.084	-.348	-.459	-.076	.057	-.245	-.337	-.392
	<i>Right side</i>									
.025	-1.073	-.047	.403	.619	.742	-.629	-.578	.282	.570	.849
.075	-.506	-.143	.175	.382	.553	-.592	-.641	-.110	.164	.442
.150	-.396	-.182	.059	.243	.408	-.607	-.517	-.225	-.003	.226
.250	-.399	-.280	-.068	.102	.265					
.350	-.425	-.361	-.182	-.032	.131					
.450	-.408	-.411	-.269	-.140	.008					
.550	-.338	-.367	-.294	-.197	-.078					
.650	-.254	-.251	-.242	-.208	-.128	-.246	-.183	-.197	-.217	-.192
.750	-.171	-.128	-.152	-.173	-.145	-.286	-.068	-.165	-.219	-.214
.850	-.067	.005	-.042	-.112	-.130	-.238	.040	-.084	-.158	-.182
.900	-.024	.053	-.013	-.114	-.162	-.200	.080	-.071	-.161	-.194

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(h)  $\alpha = 0^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.752	.687	.585	.301	-.120	.553	.794	.603	.203	-.240
.025	.282	-.058	-.510	-1.037	-.993	.334	-.072	-1.051	-1.512	-.824
.075	.149	-.101	-.375	-.834	-.833	.153	-.132	-.424	-1.320	-.627
.150	.034	-.154	-.345	-.604	-.736	.053	-.152	-.348	-.541	-.651
.250	.004	-.122	-.236	-.271	-.604	-.034	-.188	-.306	-.344	-.656
.350	-.054	-.155	-.209	-.226	-.526	-.121	-.248	-.313	-.306	-.610
.450	-.124	-.195	-.219	-.219	-.491	-.214	-.322	-.315	-.272	-.543
.550	-.147	-.196	-.176	-.178	-.421	-.247	-.338	-.266	-.235	-.474
.650	-.150	-.175	-.127	-.146	-.340	-.244	-.275	-.204	-.221	-.404
.750	-.108	-.094	-.057	-.110	-.268	-.173	-.136	-.127	-.189	-.334
.850	-.037	-.005	.014	-.048	-.183	-.063	-.012	-.035	-.120	-.277
.900	-.001	.046	.044	-.010	-.136	-.007	.042	.014	-.074	-.246
	<i>Right side</i>									
.025	-.386	.009	.334	.601	.830	-.943	-.051	.370	.601	.754
.075	-.398	-.105	.139	.355	.563	-.467	-.122	.159	.365	.543
.150	-.344	-.141	.048	.219	.393	-.351	-.146	.063	.229	.392
.250	-.247	-.126	.011	.146	.284	-.301	-.179	-.024	.120	.270
.350	-.224	-.161	-.058	.052	.173	-.313	-.232	-.102	.031	.162
.450	-.227	-.201	-.124	-.034	.070	-.281	-.268	-.170	-.074	.037
.550	-.181	-.192	-.140	-.068	.026	-.273	-.315	-.230	-.132	-.024
.650	-.126	-.151	-.133	-.077	-.011	-.201	-.251	-.214	-.150	-.067
.750	-.054	-.079	-.095	-.068	-.031	-.107	-.112	-.137	-.120	-.078
.850	.022	.016	-.015	-.014	-.013	-.016	.010	-.031	-.056	-.071
.900	.057	.052	.015	.003	-.021	.024	.050	-.001	-.047	-.103
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.516	.792	.565	.180	-.253	.710	.794	.615	.439	-.027
.025	.372	-.087	-1.131	-.797	-.736	.214	-.659	-.467	-.369	-.383
.075	.183	-.145	-.437	-.800	-.747	-.097	-.574	-.454	-.353	-.384
.150	.047	-.202	-.414	-.702	-.751	-.294	-.649	-.420	-.346	-.388
.250	-.081	-.285	-.349	-.502	-.776	-.276	-.619	-.407	-.353	-.397
.350	-.204	-.403	-.362	-.298	-.506					
.450	-.324	-.510	-.368	-.316	-.433					
.550	-.414	-.537	-.326	-.356	-.433					
.650	-.427	-.365	-.267	-.385	-.451	-.401	-.241	-.346	-.387	-.427
.750	-.220	-.126	-.214	-.399	-.467	-.210	-.118	-.344	-.386	-.426
.850	-.103	-.022	-.157	-.390	-.464	-.146	-.014	-.316	-.364	-.410
.900	-.070	.022	-.128	-.379	-.464	-.126	.032	-.298	-.351	-.400
	<i>Right side</i>									
.025	-1.094	-.041	.407	.830	.756	-.421	-.523	.288	.592	.867
.075	-.397	-.135	.180	.394	.564	-.411	-.687	-.107	.182	.452
.150	-.394	-.172	.065	.252	.419	-.428	-.594	-.219	.002	.236
.250	-.370	-.282	-.067	.107	.273					
.350	-.381	-.405	-.204	-.035	.132					
.450	-.381	-.487	-.313	-.159	-.004					
.550	-.327	-.526	-.368	-.248	-.107					
.650	-.258	-.332	-.377	-.283	-.168	-.338	-.248	-.332	-.330	-.231
.750	-.198	-.132	-.223	-.241	-.191	-.333	-.116	-.213	-.235	-.243
.850	-.126	.009	-.075	-.137	-.151	-.304	.016	-.113	-.162	-.194
.900	-.090	.060	-.044	-.137	-.180	-.278	.065	-.102	-.166	-.204



TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(i)  $\alpha = 0^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.780	.693	.638	.407	.018	.603	.808	.664	.327	-.137
.025	.300	-.042	-.458	-.958	-.986	.349	-.060	-.944	-1.309	-.588
.075	.167	-.085	-.367	-.766	-.806	.172	-.129	-.499	-1.170	-.599
.150	.053	-.146	-.354	-.628	-.700	.074	-.149	-.323	-1.038	-.620
.250	.027	-.117	-.218	-.368	-.573	-.010	-.175	-.284	-.217	-.631
.350	-.040	-.149	-.202	-.188	-.502	-.106	-.238	-.307	-.291	-.588
.450	-.115	-.208	-.215	-.194	-.472	-.205	-.323	-.304	-.247	-.529
.550	-.146	-.220	-.174	-.170	-.414	-.251	-.370	-.273	-.223	-.464
.650	-.178	-.220	-.132	-.148	-.351	-.286	-.388	-.203	-.225	-.409
.750	-.158	-.134	-.054	-.117	-.284	-.281	-.180	-.120	-.209	-.343
.850	-.071	-.011	.022	-.054	-.208	-.122	-.002	-.027	-.150	-.291
.900	-.012	.044	.058	-.018	-.161	-.033	.051	.021	-.096	-.254
<i>Right side</i>										
.025	-.337	.021	.350	.614	.847	-.896	-.038	.383	.602	.769
.075	-.401	-.094	.156	.365	.580	-.561	-.119	.175	.374	.556
.150	-.349	-.134	.060	.233	.412	-.352	-.139	.076	.241	.409
.250	-.235	-.123	.025	.156	.302	-.271	-.169	-.005	.132	.280
.350	-.216	-.158	-.045	.062	.187	-.301	-.224	-.084	.043	.176
.450	-.219	-.211	-.121	-.030	.084	-.266	-.278	-.164	-.068	.051
.550	-.173	-.224	-.151	-.076	.023	-.275	-.351	-.242	-.143	-.024
.650	-.121	-.204	-.160	-.099	-.017	-.194	-.378	-.283	-.185	-.079
.750	-.051	-.121	-.144	-.116	-.058	-.103	-.158	-.223	-.188	-.109
.850	.030	.002	-.037	-.053	-.044	-.014	.008	-.078	-.126	-.117
.900	.065	.044	-.001	-.033	-.063	.025	.049	-.023	-.103	-.147
$z/b_v = 0.66$										
<i>Left side</i>										
.000	.564	.809	.622	.280	-.148	.749	.836	.681	.493	.057
.025	.390	-.064	-1.045	-.805	-.693	.246	-.543	-.652	-.348	-.373
.075	.205	-.122	-.821	-.786	-.697	-.052	-.536	-.355	-.334	-.374
.150	.073	-.176	-.323	-.690	-.702	-.263	-.609	-.343	-.330	-.374
.250	-.047	-.250	-.342	-.452	-.758	-.252	-.635	-.346	-.334	-.383
.350	-.178	-.380	-.308	-.311	-.538					
.450	-.313	-.507	-.367	-.322	-.420					
.550	-.389	-.592	-.316	-.352	-.412					
.650	-.483	-.588	-.249	-.379	-.428	-.774	-.280	-.320	-.367	-.408
.750	-.508	-.104	-.205	-.394	-.441	-.591	-.207	-.321	-.369	-.409
.850	-.212	.000	-.160	-.391	-.447	-.199	-.134	-.303	-.352	-.404
.900	-.088	.031	-.143	-.387	-.451	-.161	-.102	-.285	-.344	-.398
<i>Right side</i>										
.025	-1.022	-.021	.430	.639	.771	-.299	-.447	.323	.621	.882
.075	-.768	-.118	.207	.407	.586	-.326	-.647	-.074	.207	.481
.150	-.299	-.150	.096	.268	.440	-.333	-.577	-.185	.029	.261
.250	-.385	-.247	-.040	.125	.294					
.350	-.310	-.384	-.175	-.018	.151					
.450	-.362	-.497	-.288	-.152	.010					
.550	-.307	-.559	-.417	-.250	-.106					
.650	-.247	-.592	-.445	-.337	-.188	-.315	-.343	-.707	-.643	-.471
.750	-.200	-.142	-.477	-.379	-.283	-.313	-.269	-.454	-.627	-.482
.850	-.137	.017	-.150	-.324	-.254	-.299	-.157	-.179	-.194	-.191
.900	-.114	.056	-.062	-.220	-.223	-.281	-.096	-.148	-.155	-.179

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(j)  $\alpha = 0^\circ$ ;  $M = 0.92$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.787	.699	.656	.452	.058	.625	.818	.680	.375	-.112
.025	.302	-.033	-.454	-.925	-.963	.348	-.054	-.930	-1.241	-.617
.075	.170	-.076	-.385	-.738	-.812	.172	-.119	-.672	-1.119	-.628
.150	.057	-.140	-.365	-.625	-.718	.076	-.136	-.391	-1.012	-.649
.250	.028	-.108	-.234	-.498	-.601	-.002	-.165	-.281	-.497	-.650
.350	-.034	-.143	-.212	-.265	-.528	-.096	-.225	-.319	-.126	-.605
.450	-.119	-.198	-.231	-.163	-.483	-.199	-.301	-.304	-.210	-.544
.550	-.150	-.214	-.184	-.155	-.414	-.250	-.361	-.291	-.233	-.483
.650	-.195	-.228	-.142	-.155	-.357	-.313	-.407	-.220	-.247	-.415
.750	-.202	-.157	-.063	-.131	-.291	-.313	-.267	-.127	-.228	-.357
.850	-.109	-.005	.015	-.072	-.226	-.209	.011	-.035	-.178	-.316
.900	-.039	.052	.052	-.037	-.186	-.078	.068	.014	-.130	-.284
	<i>Right side</i>									
.025	-.322	.033	.350	.615	.845	-.885	-.026	.379	.603	.772
.075	-.404	-.083	.159	.369	.578	-.628	-.103	.175	.375	.558
.150	-.373	-.124	.063	.232	.411	-.362	-.127	.075	.245	.410
.250	-.247	-.113	.022	.156	.302	-.266	-.154	-.004	.134	.282
.350	-.219	-.152	-.047	.060	.187	-.295	-.214	-.089	.042	.176
.450	-.227	-.201	-.123	-.032	.082	-.280	-.272	-.171	-.072	.050
.550	-.181	-.226	-.160	-.083	.022	-.277	-.353	-.246	-.150	-.027
.650	-.124	-.214	-.172	-.117	-.028	-.202	-.393	-.295	-.202	-.090
.750	-.055	-.149	-.186	-.159	-.082	-.102	-.238	-.295	-.233	-.130
.850	.025	.000	-.073	-.105	-.073	-.017	.014	-.129	-.197	-.153
.900	.059	.052	-.023	-.084	-.098	.020	.061	-.058	-.184	-.198
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.586	.822	.644	.316	-.130	.770	.852	.700	.514	.086
.025	.397	-.053	-1.031	-1.054	-.728	.258	-.535	-.784	-.351	-.393
.075	.220	-.109	-.840	-.946	-.740	-.039	-.520	-.335	-.341	-.389
.150	.086	-.164	-.312	-.674	-.760	-.244	-.594	-.335	-.340	-.392
.250	-.037	-.232	-.351	-.500	-.665	-.259	-.628	-.340	-.345	-.398
.350	-.172	-.361	-.277	-.302	-.471					
.450	-.312	-.489	-.360	-.334	-.431					
.550	-.391	-.583	-.356	-.357	-.437					
.650	-.484	-.602	-.246	-.376	-.455	-.772	-.275	-.317	-.371	-.423
.750	-.526	-.174	-.214	-.393	-.466	-.719	-.213	-.320	-.371	-.425
.850	-.481	.023	-.175	-.403	-.464	-.370	-.162	-.305	-.359	-.413
.900	-.248	.047	-.158	-.404	-.468	-.222	-.136	-.296	-.354	-.410
	<i>Right side</i>									
.025	-.993	.000	.431	.637	.774	-.272	-.412	.330	.630	.886
.075	-.785	-.095	.212	.410	.586	-.307	-.521	-.065	.217	.480
.150	-.263	-.133	.102	.274	.443	-.316	-.563	-.180	.042	.260
.250	-.357	-.230	-.031	.131	.297					
.350	-.287	-.361	-.171	-.013	.150					
.450	-.344	-.482	-.287	-.147	.006					
.550	-.338	-.551	-.414	-.248	-.113					
.650	-.238	-.586	-.458	-.341	-.197	-.305	-.344	-.740	-.633	-.478
.750	-.213	-.267	-.521	-.430	-.287	-.314	-.293	-.638	-.658	-.520
.850	-.154	.020	-.315	-.391	-.307	-.303	-.201	-.266	-.555	-.386
.900	-.141	.066	-.134	-.418	-.349	-.284	-.143	-.199	-.397	-.262

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(k)  $\alpha = 9.4^\circ$ ;  $M = 0.60$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.414	.802	.435	-.710	-1.180	.151	.689	.110	-.733	-1.196
.025	-.015	-.136	-.264	-.719	-2.110	.352	-.111	-.814	-1.304	-.993
.075	.027	-.136	-.340	-.530	-1.021	.172	-.139	-.501	-1.240	-.968
.150	-.012	-.162	-.335	-.434	-.633	.075	-.148	-.374	-.922	-.920
.250	-.008	-.111	-.248	-.304	-.418	.011	-.152	-.319	-.361	-.872
.350	-.031	-.120	-.231	-.265	-.347	-.040	-.173	-.317	-.308	-.726
.450	-.049	-.143	-.229	-.235	-.340	-.077	-.192	-.298	-.290	-.573
.550	-.049	-.120	-.190	-.208	-.283	-.079	-.166	-.238	-.254	-.454
.650	-.029	-.086	-.137	-.176	-.215	-.065	-.129	-.183	-.219	-.379
.750	-.012	-.047	-.086	-.112	-.160	-.031	-.070	-.107	-.167	-.315
.850	.024	.002	-.017	-.037	-.089	.020	-.008	-.015	-.080	-.242
.900	.043	.043	.020	-.009	-.053	.047	.038	.031	-.037	-.192
<i>Right side</i>										
.025	-.195	-.044	.020	.523	.813	-.701	-.044	.409	.607	.719
.075	-.340	-.139	.031	.336	.548	-.469	-.113	.195	.393	.548
.150	-.324	-.143	-.001	.228	.393	-.365	-.120	.100	.256	.409
.250	-.254	-.107	-.006	.167	.290	-.303	-.134	.029	.162	.294
.350	-.241	-.118	-.031	.096	.192	-.289	-.152	-.017	.096	.210
.450	-.224	-.139	-.047	.034	.114	-.273	-.171	-.056	.032	.125
.550	-.190	-.116	-.042	.014	.080	-.231	-.159	-.070	.000	.078
.650	-.139	-.079	-.024	.016	.066	-.167	-.111	-.045	-.002	.057
.750	-.077	-.038	.001	.020	.052	-.091	-.042	-.012	.004	.039
.850	-.003	.031	.045	.055	.071	.004	.022	.043	.041	.041
.900	.031	.054	.064	.064	.064	.034	.043	.054	.036	.014
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.006	.666	-.059	-.573	-.829	.296	.691	-.280	-.178	-.628
.025	.398	-.132	-1.128	-.975	-.692	.488	-.564	-1.188	-.464	-.450
.075	.216	-.150	-.621	-.909	-.715	.151	-.382	-1.107	-.450	-.441
.150	.105	-.175	-.469	-.722	-.639	-.036	-.359	-.987	-.445	-.445
.250	.024	-.192	-.413	-.514	-.582	-.077	-.320	-.757	-.454	-.454
.350	-.042	-.221	-.402	-.418	-.559					
.450	-.091	-.235	-.370	-.386	-.537					
.550	-.116	-.221	-.317	-.386	-.512					
.650	-.089	-.169	-.227	-.375	-.491	-.125	-.180	-.197	-.443	-.422
.750	-.045	-.093	-.132	-.349	-.468	-.063	-.086	-.114	-.418	-.411
.850	-.006	-.033	-.047	-.297	-.441	-.008	-.005	-.033	-.372	-.386
.900	.015	.002	-.008	-.258	-.432	.015	.031	.008	-.349	-.381
<i>Right side</i>										
.025	-.994	-.035	.451	.628	.694	-1.252	-.254	.580	.746	.950
.075	-.561	-.116	.234	.425	.571	-1.123	-.325	.165	.365	.600
.150	-.418	-.125	.130	.294	.447	-1.059	-.286	.017	.189	.393
.250	-.400	-.166	.045	.185	.336					
.350	-.393	-.203	-.024	.093	.237					
.450	-.358	-.219	-.072	.016	.144					
.550	-.298	-.203	-.093	-.025	.064					
.650	-.218	-.157	-.077	-.039	.018	-.169	-.129	-.072	-.082	-.043
.750	-.123	-.079	-.040	-.037	-.018	-.095	-.079	-.056	-.119	-.119
.850	-.019	.006	.029	-.018	-.043	-.003	.020	.024	-.105	-.135
.900	.027	.038	.052	-.032	-.089	.036	.052	.041	-.123	-.171

TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(i)  $\alpha = 9.6^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.370	.858	.414	-.395	-.170	.416	.737	.426	-.147	-.640
.025	-.032	-.111	-.154	-.580	-1.251	.341	-.093	-.968	-1.179	-.802
.075	.025	-.128	-.314	-.540	-.961	.166	-.136	-.493	-1.099	-.813
.150	-.006	-.163	-.354	-.483	-.733	.080	-.142	-.360	-.900	-.722
.250	-.007	-.107	-.248	-.329	-.493	.013	-.159	-.302	-.520	-.690
.350	-.038	-.125	-.233	-.279	-.402	-.053	-.199	-.299	-.333	-.633
.450	-.067	-.162	-.233	-.252	-.361	-.107	-.235	-.291	-.283	-.552
.550	-.067	-.142	-.199	-.218	-.316	-.107	-.206	-.234	-.255	-.480
.650	-.047	-.108	-.156	-.189	-.263	-.095	-.163	-.186	-.236	-.416
.750	-.024	-.061	-.091	-.145	-.201	-.053	-.082	-.114	-.196	-.364
.850	.017	.010	-.022	-.071	-.134	.013	.005	-.020	-.119	-.304
.900	.049	.050	.024	-.028	-.080	.051	.050	.032	-.068	-.249
	<i>Right side</i>									
.025	-.095	-.016	.000	.415	.789	-1.055	-.039	.401	.611	.736
.075	-.333	-.130	.044	.331	.540	-.452	-.102	.192	.392	.561
.150	-.342	-.143	.020	.234	.395	-.357	-.120	.100	.266	.428
.250	-.259	-.105	-.002	.174	.296	-.305	-.137	.037	.169	.316
.350	-.253	-.127	-.036	.095	.196	-.303	-.171	-.022	.095	.228
.450	-.250	-.154	-.062	.023	.113	-.305	-.200	-.079	.018	.136
.550	-.207	-.136	-.059	-.002	.084	-.265	-.192	-.096	-.024	.081
.650	-.158	-.091	-.039	-.005	.058	-.191	-.143	-.080	-.034	.043
.750	-.087	-.052	-.019	-.007	.043	-.092	-.064	-.036	-.025	.028
.850	.000	.028	.041	.032	.055	.006	.027	.027	.012	.020
.900	.039	.056	.054	.035	.045	.043	.054	.041	-.001	-.017
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.298	.696	.306	-.085	-.610	.600	.757	.418	.129	-.380
.025	.391	-.105	-1.085	-.655	-.680	.420	-.620	-.771	-.390	-.392
.075	.221	-.130	-.636	-.685	-.745	.108	-.444	-.465	-.375	-.395
.150	.111	-.166	-.391	-.654	-.604	-.081	-.416	-.468	-.378	-.402
.250	.022	-.208	-.331	-.512	-.548	-.113	-.364	-.460	-.384	-.414
.350	-.063	-.269	-.333	-.410	-.507					
.450	-.138	-.304	-.336	-.372	-.486					
.550	-.176	-.286	-.306	-.381	-.472					
.650	-.142	-.203	-.240	-.390	-.460	-.184	-.166	-.314	-.410	-.410
.750	-.083	-.097	-.173	-.381	-.448	-.103	-.050	-.293	-.400	-.395
.850	-.017	-.009	-.096	-.350	-.423	-.049	.037	-.253	-.372	-.370
.900	.006	.034	-.060	-.329	-.411	-.027	.074	-.225	-.353	-.352
	<i>Right side</i>									
.025	-1.038	-.016	.450	.628	.733	-.831	-.313	.495	.762	.995
.075	-.547	-.094	.244	.436	.605	-.708	-.384	.112	.374	.634
.150	-.379	-.114	.144	.314	.490	-.728	-.329	-.017	.194	.419
.250	-.383	-.179	.044	.207	.374					
.350	-.411	-.245	-.046	.092	.252					
.450	-.394	-.277	-.119	-.005	.149					
.550	-.325	-.266	-.154	-.061	.061					
.650	-.230	-.197	-.140	-.088	.001	-.231	-.140	-.143	-.139	-.066
.750	-.133	-.091	-.086	-.093	-.043	-.191	-.056	-.120	-.165	-.134
.850	-.021	.022	-.006	-.064	-.058	-.135	.050	-.057	-.138	-.145
.900	.022	.060	.018	-.078	-.099	-.095	.088	-.048	-.155	-.167

TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(m)  $\alpha = 9.7^\circ$ ;  $M = 0.85$

$\frac{x}{C_V}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_V = 0.11$					$z/b_V = 0.38$				
	<i>Left side</i>									
.000	.349	.879	.420	-.236	-.046	.478	.748	.500	-.009	-.513
.025	-.050	-.108	-.109	-.508	-1.154	.339	-.099	-1.172	-1.301	-.749
.075	.025	-.127	-.311	-.523	-.910	.167	-.144	-.441	-1.072	-.761
.150	-.006	-.171	-.381	-.525	-.723	.076	-.147	-.354	-.808	-.722
.250	-.013	-.117	-.256	-.341	-.536	.013	-.164	-.300	-.526	-.696
.350	-.049	-.138	-.237	-.281	-.457	-.059	-.221	-.300	-.345	-.651
.450	-.079	-.181	-.244	-.256	-.428	-.125	-.272	-.303	-.273	-.591
.550	-.076	-.165	-.207	-.216	-.370	-.132	-.249	-.254	-.241	-.526
.650	-.071	-.140	-.165	-.187	-.328	-.125	-.204	-.200	-.229	-.468
.750	-.043	-.084	-.105	-.153	-.260	-.079	-.111	-.126	-.199	-.409
.850	.010	-.005	-.027	-.083	-.179	-.005	-.002	-.031	-.136	-.341
.900	.043	.048	.022	-.042	-.135	.037	.051	.025	-.088	-.298
	<i>Right side</i>									
.025	-.042	-.007	-.030	.344	.743	-1.010	-.032	.394	.615	.727
.075	-.320	-.127	.036	.307	.497	-.437	-.104	.189	.402	.542
.150	-.347	-.147	.018	.236	.352	-.346	-.121	.095	.274	.406
.250	-.254	-.111	-.004	.182	.265	-.294	-.141	.028	.179	.297
.350	-.248	-.133	-.046	.096	.173	-.295	-.184	-.031	.103	.206
.450	-.252	-.172	-.077	.024	.087	-.298	-.225	-.095	.014	.109
.550	-.211	-.161	-.077	-.006	.057	-.267	-.229	-.123	-.035	.054
.650	-.154	-.118	-.060	-.018	.024	-.191	-.181	-.109	-.053	.014
.750	-.093	-.067	-.037	-.022	.005	-.096	-.084	-.064	-.045	-.011
.850	-.005	.020	.029	.018	.014	.003	.025	.010	-.012	-.024
.900	.035	.052	.045	.028	-.004	.038	.054	.026	-.022	-.072
	$z/b_V = 0.66$					$z/b_V = 0.93$				
	<i>Left side</i>									
.000	.364	.715	.377	-.022	-.539	.669	.783	.502	.214	-.331
.025	.389	-.108	-1.208	-.676	-.690	.401	-.639	-.709	-.348	-.425
.075	.220	-.130	-.587	-.707	-.729	.091	-.474	-.391	-.337	-.425
.150	.109	-.170	-.381	-.669	-.687	-.106	-.447	-.391	-.343	-.431
.250	.018	-.225	-.321	-.515	-.713	-.125	-.362	-.394	-.351	-.445
.350	-.079	-.309	-.318	-.364	-.568					
.450	-.169	-.359	-.329	-.333	-.506					
.550	-.227	-.392	-.308	-.353	-.494					
.650	-.194	-.244	-.246	-.371	-.494	-.227	-.174	-.313	-.384	-.458
.750	-.119	-.110	-.187	-.380	-.491	-.142	-.056	-.308	-.384	-.444
.850	-.046	-.007	-.128	-.370	-.476	-.086	.038	-.286	-.365	-.421
.900	-.022	.038	-.095	-.355	-.467	-.072	.076	-.263	-.355	-.412
	<i>Right side</i>									
.025	-1.189	.001	.446	.639	.724	-.599	-.313	.485	.770	.998
.075	-.463	-.087	.242	.448	.595	-.512	-.423	.102	.383	.621
.150	-.353	-.107	.146	.328	.478	-.522	-.343	-.027	.202	.401
.250	-.346	-.185	.039	.217	.361					
.350	-.369	-.268	-.063	.096	.234					
.450	-.374	-.329	-.148	-.008	.118					
.550	-.311	-.325	-.201	-.085	.022					
.650	-.234	-.239	-.191	-.120	-.049	-.280	-.138	-.190	-.179	-.134
.750	-.154	-.107	-.126	-.127	-.095	-.260	-.050	-.155	-.199	-.196
.850	-.058	.028	-.031	-.090	-.102	-.225	.062	-.086	-.162	-.192
.900	-.016	.072	-.011	-.109	-.151	-.194	.099	-.086	-.179	-.221

TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_1 = -6^\circ$  - Continued

(n)  $\alpha = 9.7^\circ$ ;  $M = 0.90$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.306	.964	.440	-.126	.080	.534	.750	.563	.170	-.373
.025	-.079	-.042	.020	-.393	-.792	.339	-.046	-.976	-1.290	-.621
.075	.025	-.071	-.187	-.430	-.797	.170	-.097	-.448	-1.026	-.620
.150	.003	-.130	-.347	-.526	-.640	.080	-.109	-.373	-.738	-.626
.250	-.009	-.097	-.256	-.390	-.506	.011	-.135	-.279	-.497	-.617
.350	-.052	-.120	-.236	-.292	-.450	-.064	-.195	-.286	-.392	-.583
.450	-.091	-.183	-.257	-.281	-.424	-.145	-.271	-.322	-.307	-.535
.550	-.100	-.190	-.236	-.235	-.358	-.170	-.304	-.303	-.259	-.483
.650	-.104	-.204	-.206	-.209	-.319	-.190	-.309	-.243	-.244	-.434
.750	-.086	-.132	-.140	-.168	-.259	-.150	-.173	-.159	-.227	-.382
.850	-.022	-.026	-.060	-.109	-.175	-.051	-.013	-.062	-.172	-.325
.900	.018	.032	-.009	-.067	-.127	.007	.054	-.008	-.125	-.287
	<i>Right side</i>									
.025	.043	.055	-.027	.284	.760	-.954	.013	.400	.614	.754
.075	-.272	-.073	.046	.326	.516	-.532	-.065	.204	.405	.571
.150	-.395	-.113	.052	.276	.385	-.364	-.088	.114	.284	.436
.250	-.268	-.088	.033	.200	.295	-.276	-.116	.042	.184	.331
.350	-.255	-.120	-.014	.109	.204	-.295	-.173	-.025	.101	.237
.450	-.268	-.177	-.068	.018	.116	-.309	-.237	-.104	.005	.143
.550	-.236	-.199	-.091	-.021	.084	-.295	-.274	-.154	-.056	.075
.650	-.180	-.170	-.098	-.045	.048	-.212	-.269	-.181	-.092	.033
.750	-.115	-.123	-.094	-.062	.024	-.114	-.158	-.163	-.104	.004
.850	-.024	-.013	-.032	-.028	.024	-.018	-.002	-.074	-.084	-.015
.900	.017	.028	-.005	-.024	.007	.019	.044	-.042	-.087	-.056
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.422	.721	.427	.080	-.452	.718	.815	.558	.293	-.240
.025	.394	-.053	-1.106	-.741	-.666	.404	-.479	-.754	-.378	-.416
.075	.230	-.084	-.911	-.746	-.704	.088	-.430	-.358	-.349	-.388
.150	.120	-.123	-.256	-.681	-.660	-.112	-.466	-.355	-.349	-.392
.250	.021	-.178	-.269	-.541	-.644	-.114	-.421	-.361	-.357	-.409
.350	-.084	-.299	-.304	-.344	-.508					
.450	-.196	-.396	-.351	-.342	-.459					
.550	-.298	-.443	-.339	-.367	-.453					
.650	-.310	-.445	-.265	-.381	-.454	-.574	-.244	-.317	-.385	-.419
.750	-.256	-.123	-.217	-.392	-.456	-.193	-.151	-.314	-.381	-.411
.850	-.100	.003	-.170	-.384	-.442	-.117	-.050	-.299	-.368	-.393
.900	-.059	.052	-.143	-.371	-.442	-.106	.007	-.284	-.355	-.382
	<i>Right side</i>									
.025	-1.121	.038	.453	.639	.760	-.393	-.274	.499	.769	1.012
.075	-.829	-.047	.261	.455	.623	-.391	-.441	.116	.381	.645
.150	-.271	-.073	.169	.336	.505	-.399	-.361	-.010	.200	.423
.250	-.301	-.160	.058	.225	.391					
.350	-.333	-.262	-.052	.099	.266					
.450	-.375	-.336	-.162	-.017	.153					
.550	-.324	-.408	-.240	-.118	.046					
.650	-.243	-.416	-.312	-.179	-.029	-.303	-.254	-.566	-.441	-.132
.750	-.188	-.206	-.322	-.250	-.086	-.301	-.156	-.431	-.417	-.188
.850	-.104	.029	-.170	-.205	-.089	-.281	-.029	-.136	-.234	-.179
.900	-.068	.074	-.093	-.185	-.131	-.252	.038	-.115	-.205	-.208

TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $\alpha_t = -6^\circ$  - Continued

(o)  $\alpha = 9.7^\circ$ ;  $M = 0.92$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.312	.971	.453	-.173		.569	.761	.588	.176	
.025	-.042	-.025	.044	-.371		.356	-.029	-.959	-1.302	
.075	.058	-.054	-.166	-.415		.190	-.079	-.402	-1.039	
.150	.045	-.106	-.325	-.516		.097	-.097	-.337	-.737	
.250	.021	-.068	-.227	-.391		.025	-.121	-.271	-.502	
.350	-.024	-.106	-.219	-.297		-.050	-.182	-.287	-.390	
.450	-.073	-.168	-.245	-.279		-.138	-.258	-.317	-.301	
.550	-.094	-.177	-.224	-.238		-.167	-.296	-.291	-.256	
.650	-.105	-.188	-.194	-.211		-.200	-.291	-.224	-.248	
.750	-.094	-.121	-.127	-.172		-.175	-.167	-.139	-.229	
.850	-.033	-.016	-.042	-.115		-.067	.001	-.042	-.175	
.900	.010	.037	.003	-.072		-.007	.056	.015	-.130	
<i>Right side</i>										
.025	.124	.069	.003	.282		-.855	.023	.413	.614	
.075	-.178	-.060	.073	.333		-.430	-.054	.217	.405	
.150	-.307	-.095	.068	.285		-.323	-.077	.120	.281	
.250	-.225	-.071	.039	.199		-.257	-.108	.054	.180	
.350	-.235	-.109	-.010	.106		-.285	-.157	-.017	.095	
.450	-.257	-.164	-.064	.019		-.300	-.226	-.094	.002	
.550	-.237	-.181	-.086	-.024		-.306	-.262	-.144	-.060	
.650	-.183	-.160	-.089	-.053		-.227	-.260	-.162	-.102	
.750	-.120	-.117	-.081	-.076		-.117	-.144	-.137	-.118	
.850	-.020	.001	-.017	-.039		-.017	.012	-.045	-.093	
.900	.018	.033	.010	-.035		.022	.050	-.015	-.101	
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.443	.731	.466	.088		.721	.822	.583	.303	
.025	.399	-.039	-1.076	-.742		.399	-.456	-.768	-.382	
.075	.240	-.070	-.847	-.740		.091	-.416	-.349	-.354	
.150	.133	-.110	-.264	-.682		-.107	-.454	-.350	-.358	
.250	.028	-.171	-.268	-.569		-.102	-.413	-.357	-.358	
.350	-.078	-.290	-.295	-.335						
.450	-.183	-.388	-.341	-.338						
.550	-.293	-.437	-.317	-.363						
.650	-.345	-.435	-.250	-.382		-.619	-.223	-.299	-.387	
.750	-.307	-.116	-.194	-.396		-.315	-.138	-.300	-.390	
.850	-.129	.012	-.147	-.392		-.117	-.042	-.287	-.375	
.900	-.067	.052	-.116	-.380		-.100	.001	-.266	-.367	
<i>Right side</i>										
.025	-1.049	.042	.460	.641		-.436	-.270	.488	.765	
.075	-.559	-.038	.267	.448		-.389	-.435	.112	.373	
.150	-.265	-.066	.167	.333		-.395	-.355	-.017	.190	
.250	-.310	-.152	.065	.223						
.350	-.319	-.260	-.048	.080						
.450	-.381	-.332	-.158	.025						
.550	-.348	-.405	-.236	-.128						
.650	-.233	-.412	-.300	-.190		-.291	-.268	-.555	-.456	
.750	-.181	-.203	-.290	-.263		-.291	-.155	-.323	-.478	
.850	-.099	.033	-.118	-.233		-.266	-.022	-.106	-.267	
.900	-.063	.072	-.051	-.217		-.242	.035	-.091	-.222	

TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_h = -6^\circ$  - Continued

(p)  $\alpha = 15.6^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	<i>C<sub>p</sub> for -</i>					<i>C<sub>p</sub> for -</i>				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	-.351	.825	.182	.091	-.489	.262	.740	.364	-.443	-.879
.025	-.619	-.139	.278	.098	-.491	.157	-.102	-.347	-.694	-.931
.075	-.171	-.143	.014	-.091	-.417	.075	-.143	-.309	-.507	-.798
.150	-.045	-.171	-.143	-.199	-.394	.055	-.139	-.282	-.391	-.647
.250	.007	-.127	-.152	-.180	-.331	.034	-.153	-.270	-.340	-.532
.350	.027	-.112	-.179	-.199	-.293	.011	-.182	-.275	-.315	-.451
.450	.027	-.127	-.197	-.203	-.261	-.009	-.203	-.270	-.288	-.383
.550	.025	-.121	-.170	-.176	-.213	.000	-.171	-.225	-.244	-.326
.650	.034	-.086	-.132	-.135	-.168	.000	-.139	-.177	-.206	-.279
.750	.034	-.057	-.093	-.096	-.123	.009	-.089	-.116	-.149	-.218
.850	.034	-.009	-.036	-.041	-.067	.034	-.018	-.043	-.087	-.150
.900	.052	.027	-.004	-.009	-.035	.043	.023	-.002	-.046	-.107
	<i>Right side</i>									
.025	.344	-.098	-.732	-.260	-.089	-.460	-.096	.123	.196	.211
.075	-.023	-.175	-.207	-.075	.030	-.364	-.139	.050	.123	.245
.150	-.162	-.164	-.059	.004	.112	-.317	-.148	.034	.107	.256
.250	-.175	-.125	.005	.055	.161	-.285	-.150	.032	.109	.270
.350	-.200	-.121	.025	.075	.182	-.278	-.166	.021	.100	.261
.450	-.205	-.137	.023	.080	.179	-.266	-.182	.003	.087	.233
.550	-.171	-.118	.028	.080	.159	-.230	-.168	-.004	.078	.204
.650	-.130	-.082	.032	.075	.132	-.173	-.130	.009	.073	.170
.750	-.086	-.052	.037	.068	.109	-.107	-.073	.028	.071	.143
.850	-.018	.011	.057	.073	.093	-.034	.002	.053	.075	.116
.900	.009	.032	.053	.066	.075	-.004	.018	.046	.059	.084
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	-.043	.665	.062	-.530	-.760	.248	.731	-.347	-.496	-.708
.025	.453	-.100	-1.027	-1.103	-.744	.627	-.399	-1.109	-.612	-.491
.075	.276	-.121	-.577	-.998	-.724	.285	-.305	-1.100	-.601	-.462
.150	.171	-.148	-.438	-.806	-.681	.100	-.287	-.968	-.619	-.475
.250	.103	-.166	-.384	-.569	-.631	.059	-.228	-.707	-.598	-.466
.350	.043	-.203	-.372	-.452	-.579					
.450	-.002	-.221	-.354	-.395	-.520					
.550	-.036	-.212	-.297	-.352	-.460					
.650	-.023	-.162	-.218	-.288	-.387	-.082	-.178	-.188	-.386	-.329
.750	.000	-.091	-.132	-.212	-.331	-.027	-.082	-.107	-.322	-.295
.850	.021	-.032	-.054	-.137	-.272	.007	-.002	-.027	-.256	-.272
.900	.027	.000	-.009	-.112	-.243	.021	.032	.016	-.231	-.250
	<i>Right side</i>									
.025	-1.105	-.064	.455	.621	.610	-1.086	-.287	.628	.822	.960
.075	-.642	.121	.255	.422	.502	-1.052	-.278	.264	.470	.656
.150	-.430	-.123	.164	.310	.428	-.900	-.262	.103	.278	.457
.250	-.392	-.164	.100	.228	.362					
.350	-.387	-.196	.037	.148	.297					
.450	-.351	-.209	.000	.105	.240					
.550	-.296	-.203	-.022	.068	.184					
.650	-.219	-.162	-.013	.052	.148	-.168	-.091	-.002	.050	.098
.750	-.134	-.089	.009	.048	.103	-.116	-.073	-.009	-.009	.001
.850	-.032	.000	.053	.062	.078	-.032	.018	.048	.000	-.024
.900	.009	.030	.068	.055	.035	.005	.052	.062	-.016	-.053



TABLE IV.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(q)  $\alpha = 15.8^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.032	.859	.404	.126	-.499	.332	.778	.299	-.229	-.555
.025	-.955	.095	.327	.051	-.452	-.002	-.049	-.314	-.534	-.645
.075	-.574	-.097	.007	-.121	-.410	-.067	-.139	-.323	-.438	-.584
.150	-.260	-.197	-.192	-.236	-.386	-.059	-.156	-.300	-.380	-.517
.250	-.124	-.142	-.179	-.197	-.325	-.041	-.162	-.286	-.339	-.463
.350	-.061	-.123	-.197	-.204	-.278	-.041	-.197	-.289	-.318	-.413
.450	-.017	-.139	-.215	-.207	-.234	-.041	-.227	-.279	-.294	-.360
.550	.001	-.132	-.188	-.171	-.189	-.018	-.201	-.242	-.254	-.311
.650	.024	-.091	-.144	-.138	-.152	-.008	-.157	-.197	-.223	-.269
.750	.035	-.055	-.094	-.103	-.116	.009	-.095	-.127	-.168	-.214
.850	.047	-.003	-.040	-.046	-.066	.041	-.009	-.051	-.106	-.157
.900	.057	.033	-.007	-.016	-.033	.054	.036	-.009	-.066	-.122
	<i>Right side</i>									
.025	.394	-.768	-.948	-.452	-.302	-.420	-.269	-.050	.005	.055
.075	-.014	-.413	-.585	-.250	-.172	-.360	-.260	-.092	-.028	.102
.150	-.210	-.251	-.279	-.166	-.048	-.320	-.201	-.072	-.016	.164
.250	-.204	-.139	-.136	-.081	.058	-.292	-.154	-.040	.017	.220
.350	-.221	-.088	-.065	-.019	.130	-.283	-.142	-.030	.034	.242
.450	-.222	-.077	-.022	.019	.163	-.283	-.151	-.027	.045	.236
.550	-.191	-.053	.008	.043	.157	-.249	-.138	-.013	.054	.217
.650	-.141	-.021	.026	.055	.139	-.189	-.094	.007	.064	.193
.750	-.095	-.002	.037	.061	.117	-.115	-.036	.031	.067	.157
.850	-.021	.042	.057	.073	.093	-.035	.030	.061	.076	.127
.900	.000	.053	.060	.064	.078	-.003	.047	.052	.061	.090
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.364	.708	.454	.070	-.390	.625	.790	.356	-.075	-.523
.025	.380	-.061	-1.059	-.918	-.631	.529	-.497	-.934	-.508	-.499
.075	.220	-.110	-.532	-.810	-.611	.202	-.363	-.582	-.406	-.401
.150	.128	-.147	-.402	-.649	-.584	.038	-.345	-.603	-.408	-.411
.250	.066	-.186	-.347	-.468	-.542	.024	-.248	-.573	-.408	-.408
.350	.009	-.237	-.356	-.373	-.505					
.450	-.046	-.277	-.361	-.330	-.463					
.550	-.071	-.266	-.315	-.309	-.414					
.650	-.053	-.198	-.232	-.283	-.363	-.129	-.169	-.277	-.374	-.284
.750	-.018	-.097	-.151	-.251	-.314	-.059	-.047	-.239	-.348	-.257
.850	.016	-.014	-.068	-.201	-.270	-.021	.042	-.186	-.311	-.230
.900	.030	.029	-.031	-.171	-.248	-.015	.075	-.153	-.292	-.220
	<i>Right side</i>									
.025	-1.112	-.085	.400	.597	.687	-.624	-.289	.542	.799	1.013
.075	-.536	-.126	.196	.381	.557	-.610	-.302	.193	.435	.687
.150	-.373	-.109	.137	.286	.480	-.632	-.260	.051	.246	.486
.250	-.357	-.144	.072	.205	.413					
.350	-.378	-.186	.005	.125	.342					
.450	-.369	-.204	-.037	.076	.278					
.550	-.310	-.197	-.056	.049	.217					
.650	-.230	-.136	-.040	.040	.174	-.246	-.077	-.021	.028	.105
.750	-.147	-.059	-.009	.037	.128	-.225	-.024	-.039	-.034	.004
.850	-.042	.039	.048	.051	.084	-.159	.071	.010	-.025	-.022
.900	.004	.074	.061	.037	.048	-.126	.107	.010	-.050	-.054

TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Continued

(r)  $\alpha = 15.9^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.309	.904	.574	.140	-.587	.420	.775	.349	-.237	-.522
.025	-1.069	.002	.247	-.063	-.552	.097	-.153	-.435	-.595	-.646
.075	-.603	-.205	-.046	-.211	-.498	-.041	-.230	-.418	-.504	-.594
.150	-.288	-.298	-.278	-.320	-.457	-.072	-.216	-.368	-.438	-.533
.250	-.179	-.206	-.233	-.254	-.387	-.079	-.204	-.326	-.382	-.475
.350	-.130	-.164	-.235	-.240	-.322	-.098	-.235	-.318	-.348	-.432
.450	-.092	-.162	-.241	-.233	-.265	-.113	-.273	-.309	-.321	-.384
.550	-.055	-.162	-.219	-.197	-.209	-.085	-.242	-.274	-.277	-.336
.650	-.025	-.104	-.159	-.149	-.168	-.062	-.192	-.220	-.241	-.299
.750	.000	-.060	-.111	-.117	-.130	-.034	-.109	-.151	-.187	-.230
.850	.023	-.009	-.052	-.063	-.076	.006	-.018	-.069	-.126	-.188
.900	.046	.027	-.013	-.035	-.044	.030	.033	-.023	-.083	-.148
	<i>Right side</i>									
.025	.318	-.685	-1.070	-.511	-.418	-.791	-.279	-.069	-.052	-.016
.075	-.096	-.845	-.664	-.324	-.285	-.573	-.315	-.146	-.096	.014
.150	-.354	-.330	-.337	-.241	-.152	-.427	-.262	-.145	-.086	.079
.250	-.284	-.233	-.220	-.166	-.034	-.312	-.211	-.108	-.052	.142
.350	-.254	-.160	-.153	-.092	.058	-.298	-.194	-.094	-.032	.169
.450	-.238	-.126	-.094	-.038	.111	-.295	-.201	-.084	-.013	.183
.550	-.190	-.073	-.047	-.004	.124	-.266	-.178	-.066	.002	.178
.650	-.150	-.042	-.012	.021	.114	-.207	-.124	-.036	.026	.163
.750	-.099	-.015	.013	.036	.104	-.127	-.058	-.004	.039	.134
.850	-.021	.030	.043	.052	.086	-.047	.023	.036	.052	.110
.900	.002	.049	.045	.051	.066	-.011	.040	.033	.038	.068
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.368	.709	.496	.166	-.234	.680	.809	.497	.061	-.420
.025	.368	-.102	-1.170	-.941	-.621	.498	-.555	-.886	-.448	-.512
.075	.200	-.144	-.544	-.834	-.605	.172	-.422	-.469	-.374	-.402
.150	.105	-.175	-.400	-.662	-.578	-.011	-.386	-.484	-.375	-.411
.250	.037	-.215	-.336	-.489	-.540	-.017	-.239	-.479	-.380	-.415
.350	-.040	-.279	-.346	-.375	-.501					
.450	-.113	-.334	-.359	-.324	-.468					
.550	-.147	-.347	-.318	-.303	-.428					
.650	-.120	-.230	-.247	-.287	-.384	-.201	-.165	-.313	-.377	-.320
.750	-.074	-.111	-.175	-.271	-.343	-.116	-.049	-.295	-.360	-.291
.850	-.031	-.017	-.104	-.236	-.299	-.072	.040	-.257	-.337	-.257
.900	-.015	.025	-.064	-.216	-.282	-.069	.074	-.233	-.327	-.248
	<i>Right side</i>									
.025	-1.301	-.099	.359	.576	.708	-.498	-.290	.487	.780	1.006
.075	-.710	-.157	.162	.357	.551	-.486	-.381	.134	.398	.673
.150	-.320	-.143	.095	.254	.464	-.496	-.320	-.015	.204	.455
.250	-.326	-.187	.028	.170	.389					
.350	-.349	-.236	-.049	.086	.310					
.450	-.359	-.256	-.094	.039	.247					
.550	-.316	-.249	-.114	-.001	.193					
.650	-.249	-.177	-.098	-.005	.145	-.301	-.077	-.059	-.018	.094
.750	-.180	-.079	-.054	-.002	.103	-.289	-.029	-.083	-.073	-.028
.850	-.085	.034	.016	.021	.068	-.245	.074	-.030	-.056	-.051
.900	-.041	.070	.029	.001	.024	-.208	.105	-.037	-.083	-.095

TABLE IV. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = -6^\circ$  - Concluded

(s)  $\alpha = 15.9^\circ$ ;  $M = 0.90$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.560	.906	.777	.280		.462	.756	.517	.166	
.025	-1.029	-.264	.238	-.202		.349	-.138	-1.117	-1.363	
.075	-.512	-.365	-.063	-.295		.150	-.220	-.593	-.926	
.150	-.178	-.368	-.354	-.480		.049	-.240	-.567	-.754	
.250	-.138	-.339	-.605	-.658		-.027	-.226	-.607	-.698	
.350	-.145	-.246	-.497	-.584		-.110	-.269	-.295	-.420	
.450	-.161	-.234	-.229	-.280		-.186	-.344	-.293	-.404	
.550	-.209	-.324	-.319	-.324		-.212	-.388	-.346	-.353	
.650	-.197	-.229	-.231	-.268		-.230	-.340	-.286	-.297	
.750	-.164	-.144	-.163	-.214		-.198	-.198	-.191	-.244	
.850	-.105	-.054	-.083	-.146		-.120	-.050	-.101	-.188	
.900	-.060	-.003	-.030	-.092		-.176	-.194	-.175	-.154	
<i>Right side</i>										
.025	.330	-.226	-.951	-.105		-1.101	-.130	.371	.461	
.075	-.073	-.487	-.489	-.154		-.618	-.214	.145	.236	
.150	-.377	-.350	-.182	-.140		-.598	-.230	.039	.111	
.250	-.619	-.320	-.150	-.137		-.596	-.218	-.029	.035	
.350	-.491	-.261	-.165	-.136		-.319	-.265	-.091	-.032	
.450	-.273	-.289	-.189	-.149		-.284	-.319	-.166	-.101	
.550	-.241	-.241	-.189	-.118		-.332	-.333	-.203	-.130	
.650	-.218	-.205	-.186	-.112		-.284	-.320	-.213	-.140	
.750	-.172	-.150	-.158	-.088		-.188	-.198	-.183	-.124	
.850	-.072	-.040	-.073	-.044		-.090	-.042	-.093	-.081	
.900	-.038	-.006	-.047	-.038		-.052	-.006	-.065	-.096	
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.321	.695	.366	-.015		.699	.823	.499	.079	
.025	.409	-.088	-1.231	-.746		.535	-.505	-.821	-.380	
.075	.244	-.124	-1.047	-.747		.201	-.463	-.391	-.373	
.150	.131	-.160	-.622	-.748		-.006	-.391	-.393	-.375	
.250	.039	-.214	-.207	-.663		-.015	-.258	-.405	-.394	
.350	-.070	-.323	-.270	-.547						
.450	-.190	-.415	-.375	-.465						
.550	-.300	-.467	-.401	-.369						
.650	-.312	-.504	-.318	-.351		-.546	-.212	-.367	-.423	
.750	-.324	-.144	-.267	-.365		-.469	-.165	-.365	-.427	
.850	-.230	-.034	-.214	-.369		-.196	-.096	-.342	-.420	
.900	-.168	.003	-.179	-.360		-.174	-.050	-.321	-.415	
<i>Right side</i>										
.025	-1.258	-.049	.431	.600		-.421	-.216	.547	.823	
.075	-1.046	-.116	.233	.412		-.401	-.459	.179	.437	
.150	-.574	-.125	.143	.298		-.403	-.345	.009	.217	
.250	-.198	-.206	.042	.194						
.350	-.289	-.311	-.069	.062						
.450	-.373	-.377	-.174	-.050						
.550	-.387	-.440	-.262	-.145						
.650	-.319	-.439	-.323	-.200		-.363	-.309	-.491	-.293	
.750	-.276	-.288	-.338	-.222		-.369	-.218	-.334	-.269	
.850	-.205	-.015	-.153	-.152		-.349	-.081	-.138	-.202	
.900	-.176	.029	-.101	-.166		-.319	-.018	-.135	-.226	

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$

(a)  $\alpha = 0^\circ$ ;  $M = 0.60$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.642	.647	.334	-.438	-1.286	.262	.741	.289	-.688	-1.166
.025	.276	-.096	-.561	-1.134	-2.145	.346	-.098	-.647	-1.405	-1.064
.075	.141	-.105	-.372	-.672	-1.904	.160	-.131	-.443	-1.063	-1.038
.150	.048	-.133	-.307	-.500	-.677	.066	-.141	-.329	-.573	-.967
.250	.023	-.100	-.213	-.307	-.404	.002	-.138	-.279	-.403	-.907
.350	-.011	-.112	-.182	-.240	-.325	-.055	-.176	-.272	-.348	-.772
.450	-.052	-.124	-.168	-.201	-.284	-.089	-.190	-.252	-.297	-.613
.550	-.057	-.105	-.127	-.157	-.284	-.082	-.150	-.200	-.231	-.478
.650	-.039	-.074	-.088	-.113	-.219	-.066	-.115	-.145	-.171	-.383
.750	-.014	-.029	-.034	-.072	-.140	-.030	-.065	-.079	-.100	-.293
.850	.030	.025	.023	-.015	-.069	.027	.006	-.002	-.021	-.200
.900	.055	.051	.055	.020	-.022	.052	.041	.041	.025	-.150
	<i>Right side</i>									
.025	-.449	-.034	.309	.576	.797	-.645	-.065	.382	.608	.749
.075	-.389	-.115	.141	.349	.550	-.435	-.126	.171	.379	.533
.150	-.305	-.122	.062	.220	.383	-.337	-.133	.078	.238	.378
.250	-.219	-.105	.030	.149	.269	-.273	-.143	.016	.142	.246
.350	-.194	-.115	-.018	.077	.175	-.253	-.157	-.032	.071	.168
.450	-.173	-.129	-.057	.022	.100	-.241	-.162	-.070	.022	.098
.550	-.132	-.105	-.054	.006	.068	-.203	-.145	-.072	-.003	.054
.650	-.080	-.065	-.025	.015	.059	-.143	-.107	-.047	.008	.043
.750	-.030	-.027	-.002	.031	.056	-.071	-.046	-.011	.029	.038
.850	.034	.034	.050	.061	.073	.007	.030	.055	.064	.047
.900	.057	.053	.064	.071	.073	.039	.041	.062	.064	.022
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.212	.732	.239	-.451	-.640	.358	.772	.114	-.490	-.610
.025	.374	-.115	-.875	-1.146	-.691	.483	-.058	-1.002	-1.015	-.545
.075	.185	-.138	-.513	-1.086	-.668	.330	-.070	-.572	-.842	-.494
.150	.082	-.155	-.400	-.975	-.670	.180	-.129	-.475	-.787	-.497
.250	.016	-.157	-.332	-.709	-.629	.052	-.195	-.422	-.677	-.497
.350	-.036	-.181	-.311	-.467	-.580					
.450	-.080	-.200	-.297	-.359	-.545					
.550	-.102	-.200	-.268	-.302	-.506					
.650	-.080	-.157	-.209	-.238	-.464	-.143	-.235	-.293	-.346	-.381
.750	-.041	-.105	-.134	-.164	-.423	-.118	-.181	-.216	-.288	-.353
.850	-.007	-.037	-.057	-.097	-.381	-.080	-.112	-.129	-.226	-.316
.900	.009	-.006	-.016	-.061	-.363	-.050	-.074	-.086	-.189	-.300
	<i>Right side</i>									
.025	-.802	-.070	.414	.615	.702	-.779	.013	.514	.675	.735
.075	-.506	-.129	.191	.395	.519	-.526	-.051	.323	.528	.658
.150	-.362	-.138	.100	.259	.385	-.433	-.110	.203	.404	.561
.250	-.326	-.155	.028	.165	.274					
.350	-.314	-.176	-.029	.091	.188					
.450	-.296	-.190	-.061	.038	.124					
.550	-.262	-.183	-.077	.011	.070					
.650	-.205	-.152	-.066	.002	.033	-.273	-.195	-.102	-.033	-.011
.750	-.134	-.110	-.032	.008	.001	-.207	-.171	-.102	-.072	-.092
.850	-.032	.011	.025	.043	-.022	-.100	-.065	-.032	-.042	-.108
.900	.011	.027	.050	.043	-.069	-.055	-.037	-.009	-.051	-.140

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(b)  $\alpha = 0^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.725	.677	.523	.201	-.240	.475	.775	.520	.010	-.428
.025	.286	-.069	-.548	-1.042	-1.000	.347	-.069	-.983	-1.715	-.683
.075	.155	-.090	-.374	-.872	-.884	.163	-.122	-.462	-1.506	-.688
.150	.055	-.133	-.330	-.549	-.798	.071	-.141	-.352	-.482	-.702
.250	.029	-.097	-.228	-.312	-.664	.000	-.152	-.301	-.370	-.686
.350	-.015	-.111	-.196	-.246	-.539	-.061	-.174	-.290	-.349	-.616
.450	-.063	-.133	-.187	-.218	-.452	-.110	-.197	-.279	-.303	-.531
.550	-.063	-.111	-.115	-.162	-.341	-.096	-.169	-.225	-.234	-.438
.650	-.054	-.082	-.095	-.124	-.266	-.078	-.127	-.164	-.174	-.359
.750	-.020	-.027	-.030	-.071	-.199	-.032	-.060	-.085	-.096	-.297
.850	.032	.033	.035	-.009	-.116	.030	.019	.008	-.009	-.250
.900	.059	.069	.064	.031	-.064	.065	.061	.055	.037	-.213
	<i>Right side</i>									
.025	-.400	.004	.324	.593	.810	-.977	-.057	.379	.608	.752
.075	-.395	-.094	.149	.360	.560	-.442	-.115	.166	.374	.545
.150	-.313	-.115	.066	.230	.394	-.343	-.127	.076	.239	.394
.250	-.229	-.099	.028	.159	.294	-.287	-.139	.006	.144	.276
.350	-.203	-.118	-.023	.079	.190	-.272	-.161	-.042	.074	.187
.450	-.191	-.129	-.061	.016	.104	-.265	-.175	-.083	.010	.104
.550	-.144	-.113	-.064	-.003	.064	-.223	-.158	-.091	-.016	.055
.650	-.084	-.066	-.036	.005	.044	-.156	-.111	-.061	-.009	.038
.750	-.029	-.027	-.009	.019	.037	-.069	-.041	-.012	.013	.023
.850	.042	.046	.047	.059	.047	.023	.043	.052	.057	.017
.900	.074	.065	.060	.066	.040	.059	.058	.060	.056	-.023
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.440	.764	.488	-.003	-.336	.533	.811	.397	-.139	-.349
.025	.370	-.099	-1.152	-1.336	-.735	.483	-.012	-1.167	-1.093	-.402
.075	.192	-.129	-.565	-1.182	-.761	.350	-.044	-.691	-.577	-.390
.150	.081	-.150	-.422	-.950	-.773	.203	-.113	-.545	-.580	-.414
.250	.018	-.157	-.345	-.692	-.689	.065	-.202	-.425	-.589	-.423
.350	-.037	-.186	-.334	-.480	-.484					
.450	-.087	-.220	-.340	-.373	-.449					
.550	-.118	-.224	-.314	-.332	-.451					
.650	-.096	-.177	-.238	-.283	-.455	-.163	-.253	-.308	-.382	-.399
.750	-.054	-.107	-.143	-.215	-.447	-.137	-.182	-.208	-.318	-.384
.850	-.005	-.027	-.047	-.142	-.414	-.081	-.097	-.106	-.258	-.344
.900	.023	.007	-.003	-.102	-.400	-.043	-.049	-.059	-.235	-.317
	<i>Right side</i>									
.025	-1.220	-.054	.408	.624	.731	-1.160	.055	.525	.705	.782
.075	-.517	-.125	.187	.396	.545	-.601	-.024	.354	.559	.706
.150	-.371	-.122	.099	.272	.411	-.476	-.074	.230	.435	.607
.250	-.333	-.146	.025	.172	.300					
.350	-.331	-.171	-.030	.097	.212					
.450	-.330	-.206	-.068	.040	.145					
.550	-.299	-.206	-.092	.001	.078					
.650	-.226	-.171	-.079	-.016	.034	-.297	-.227	-.123	-.067	-.018
.750	-.131	-.111	-.042	-.016	-.012	-.197	-.171	-.115	-.117	-.119
.850	-.017	.023	.032	.019	-.044	-.072	-.051	-.026	-.085	-.125
.900	.035	.044	.055	.022	-.090	-.020	-.016	-.003	-.099	-.161

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(c)  $\alpha = 0^\circ$ ;  $M = 0.85$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.757	.682	.574	.327	-.098	.531	.789	.611	.166	-.292
.025	.283	-.061	-.522	-.946	-.975	.342	-.072	-1.052	-1.453	-.665
.075	.156	-.091	-.369	-.828	-.849	.160	-.131	-.424	-1.287	-.667
.150	.054	-.138	-.343	-.648	-.751	.070	-.145	-.355	-1.113	-.678
.250	.029	-.111	-.232	-.343	-.626	-.004	-.159	-.306	-.276	-.658
.350	-.020	-.122	-.202	-.241	-.535	-.066	-.189	-.302	-.287	-.607
.450	-.070	-.148	-.199	-.206	-.478	-.122	-.218	-.292	-.296	-.535
.550	-.073	-.128	-.152	-.160	-.387	-.109	-.194	-.239	-.240	-.455
.650	-.064	-.102	-.106	-.131	-.311	-.087	-.152	-.177	-.181	-.381
.750	-.027	-.045	-.034	-.077	-.240	-.040	-.078	-.087	-.110	-.306
.850	.026	.029	.036	-.013	-.153	.029	.015	.013	-.020	-.260
.900	.057	.068	.067	.029	-.108	.070	.065	.061	.033	-.224
	<i>Right side</i>									
.025	-.369	.010	.329	.596	.814	-.967	-.045	.373	.599	.744
.075	-.388	-.092	.147	.360	.555	-.441	-.118	.166	.369	.536
.150	-.322	-.116	.066	.230	.387	-.349	-.134	.074	.236	.383
.250	-.238	-.108	.027	.159	.281	-.295	-.152	.004	.137	.261
.350	-.216	-.129	-.027	.074	.181	-.282	-.175	-.052	.059	.176
.450	-.202	-.148	-.073	.003	.087	-.275	-.194	-.097	-.007	.087
.550	-.156	-.132	-.077	-.018	.047	-.236	-.181	-.104	-.036	.034
.650	-.090	-.079	-.046	-.010	.024	-.170	-.135	-.073	-.028	.014
.750	-.032	-.038	-.022	-.001	.011	-.072	-.059	-.022	-.006	-.002
.850	.043	.038	.046	.049	.024	.026	.042	.050	.040	-.012
.900	.074	.070	.064	.059	.008	.063	.062	.061	.040	-.057
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.498	.777	.554	.147	-.237	.569	.819	.458	-.021	-.290
.025	.363	-.102	-1.167	-1.411	-.743	.488	-.048	-1.153	-1.153	-.463
.075	.183	-.141	-.575	-1.214	-.762	.358	-.051	-.787	-.511	-.392
.150	.079	-.164	-.412	-.901	-.772	.205	-.114	-.579	-.508	-.415
.250	.011	-.168	-.333	-.638	-.713	.060	-.216	-.411	-.513	-.431
.350	-.046	-.205	-.341	-.450	-.477					
.450	-.093	-.245	-.358	-.346	-.439					
.550	-.133	-.259	-.332	-.324	-.442					
.650	-.113	-.213	-.258	-.298	-.451	-.187	-.289	-.311	-.413	-.408
.750	-.067	-.178	-.142	-.251	-.454	-.150	-.199	-.203	-.370	-.399
.850	-.009	-.028	-.034	-.191	-.435	-.084	-.098	-.094	-.313	-.367
.900	.020	.013	.010	-.150	-.418	-.044	-.047	-.043	-.274	-.350
	<i>Right side</i>									
.025	-1.150	-.047	.403	.610	.727	-1.206	.069	.524	.703	.792
.075	-.435	-.124	.190	.387	.537	-.630	-.019	.353	.563	.707
.150	-.365	-.126	.097	.263	.399	-.482	-.077	.230	.432	.603
.250	-.329	-.162	.024	.163	.289					
.350	-.342	-.195	-.039	.089	.200					
.450	-.353	-.226	-.086	.029	.122					
.550	-.343	-.241	-.109	-.023	.057					
.650	-.253	-.202	-.097	-.043	.002	-.306	-.251	-.136	-.098	-.056
.750	-.140	-.125	-.057	-.047	-.047	-.196	-.182	-.132	-.156	-.156
.850	-.013	.012	.024	-.014	-.067	-.062	-.051	-.037	-.123	-.159
.900	.040	.045	.053	-.014	-.122	-.012	-.011	-.009	-.133	-.199

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(d)  $\alpha = 0^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.786	.693	.619	.436	.052	.585	.798	.670	.300	-.158
.025	.297	-.050	-.472	-.861	-.844	.345	-.069	-.935	-1.268	-.607
.075	.169	-.084	-.377	-.747	-.732	.165	-.138	-.595	-1.141	-.614
.150	.063	-.144	-.362	-.648	-.688	.073	-.155	-.393	-1.036	-.632
.250	.033	-.117	-.252	-.489	-.610	.002	-.172	-.324	-.591	-.619
.350	-.015	-.136	-.222	-.317	-.532	-.065	-.206	-.343	-.321	-.575
.450	-.074	-.168	-.223	-.222	-.470	-.124	-.244	-.301	-.166	-.516
.550	-.078	-.147	-.167	-.133	-.394	-.116	-.227	-.274	-.182	-.461
.650	-.074	-.125	-.124	-.115	-.334	-.096	-.187	-.214	-.175	-.395
.750	-.031	-.061	-.043	-.069	-.265	-.048	-.103	-.102	-.119	-.319
.850	.030	.021	.031	-.010	-.195	.033	.011	.010	-.033	-.273
.900	.064	.062	.070	.027	-.146	.076	.068	.067	.019	-.232
	<i>Right side</i>									
.025	-.324	.024	.331	.602	.833	-.893	-.042	.364	.595	.760
.075	-.393	-.089	.151	.368	.575	-.542	-.125	.162	.367	.552
.150	-.330	-.123	.065	.233	.409	-.373	-.142	.065	.240	.403
.250	-.241	-.115	.022	.159	.304	-.296	-.163	-.007	.138	.284
.350	-.222	-.143	-.035	.072	.194	-.297	-.194	-.064	.059	.190
.450	-.207	-.166	-.087	.000	.093	-.276	-.217	-.113	-.013	.096
.550	-.159	-.156	-.099	-.028	.048	-.255	-.210	-.126	-.045	.041
.650	-.092	-.101	-.063	-.021	.015	-.185	-.171	-.099	-.048	.011
.750	-.027	-.053	-.032	-.016	-.002	-.077	-.084	-.035	-.025	-.007
.850	.052	.033	.038	.035	-.002	.035	.034	.045	.016	-.025
.900	.083	.066	.057	.043	-.021	.076	.062	.059	.022	-.073
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.551	.785	.618	.288	-.123	.616	.833	.534	.117	-.202
.025	.362	-.100	-1.051	-1.350	-.716	.481	-.030	-.970	-1.091	-.410
.075	.188	-.146	-.869	-1.211	-.727	.364	-.045	-.903	-.449	-.366
.150	.080	-.174	-.394	-1.075	-.734	.217	-.109	-.635	-.460	-.387
.250	.014	-.181	-.392	-.620	-.692	.063	-.214	-.515	-.458	-.402
.350	-.044	-.217	-.272	-.357	-.430					
.450	-.101	-.276	-.383	-.281	-.409					
.550	-.140	-.305	-.440	-.294	-.413					
.650	-.130	-.273	-.312	-.299	-.422	-.204	-.409	-.425	-.410	-.383
.750	-.078	-.174	-.171	-.270	-.432	-.162	-.213	-.203	-.394	-.379
.850	-.009	-.021	-.028	-.224	-.425	-.084	-.084	-.075	-.355	-.359
.900	.026	.025	.024	-.194	-.415	-.034	-.032	-.027	-.330	-.346
	<i>Right side</i>									
.025	-1.036	-.050	.390	.602	.742	-1.057	.071	.510	.706	.815
.075	-.805	-.135	.178	.384	.554	-.847	-.025	.351	.566	.728
.150	-.346	-.132	.089	.264	.421	-.476	-.077	.228	.440	.619
.250	-.352	-.163	.011	.162	.309					
.350	-.306	-.207	-.050	.080	.221					
.450	-.370	-.252	-.101	.019	.144					
.550	-.349	-.285	-.141	-.039	.068					
.650	-.320	-.268	-.132	-.067	.009	-.379	-.367	-.182	-.133	-.053
.750	-.139	-.158	-.083	-.081	-.049	-.185	-.201	-.151	-.207	-.166
.850	.006	.014	.019	-.043	-.074	-.046	-.050	-.039	-.159	-.162
.900	.058	.057	.049	-.051	-.125	.006	-.002	-.008	-.184	-.214

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(e)  $\alpha = 0^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.787	.691	.634	.466	.106	.614	.802	.687	.334	-.109
.025	.283	-.054	-.443	-.845	-.854	.325	-.077	-.904	-1.209	-.603
.075	.156	-.090	-.367	-.722	-.744	.148	-.150	-.670	-1.097	-.610
.150	.047	-.155	-.363	-.639	-.692	.055	-.170	-.412	-1.002	-.630
.250	.018	-.132	-.271	-.530	-.613	-.018	-.185	-.327	-.643	-.626
.350	-.032	-.155	-.231	-.354	-.532	-.084	-.226	-.351	-.498	-.580
.450	-.096	-.192	-.239	-.276	-.469	-.150	-.269	-.356	-.156	-.515
.550	-.105	-.180	-.179	-.139	-.395	-.150	-.253	-.247	-.139	-.461
.650	-.108	-.164	-.130	-.105	-.335	-.134	-.241	-.237	-.160	-.399
.750	-.057	-.086	-.050	-.067	-.273	-.074	-.142	-.112	-.122	-.326
.850	.010	.007	.033	-.012	-.207	.013	-.001	.008	-.041	-.276
.900	.049	.054	.069	.021	-.164	.063	.062	.069	.012	-.244
	<i>Right side</i>									
.025	-.302	.020	.337	.603	.831	-.830	-.045	.361	.594	.763
.075	-.390	-.090	.153	.366	.575	-.609	-.134	.161	.365	.550
.150	-.356	-.129	.066	.237	.409	-.409	-.154	.063	.234	.401
.250	-.272	-.129	.021	.159	.304	-.317	-.177	-.009	.132	.279
.350	-.245	-.162	-.039	.070	.192	-.335	-.213	-.070	.049	.196
.450	-.238	-.189	-.093	-.002	.087	-.325	-.243	-.122	-.023	.092
.550	-.183	-.184	-.106	-.039	.043	-.250	-.236	-.142	-.058	.034
.650	-.111	-.126	-.071	-.035	.004	-.230	-.215	-.114	-.060	.004
.750	-.046	-.079	-.045	-.033	-.012	-.103	-.126	-.045	-.038	-.016
.850	.038	.019	.030	.020	-.010	.021	.021	.038	.005	-.035
.900	.073	.053	.052	.032	-.036	.065	.053	.055	.008	-.083
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.577	.786	.640	.321	-.077	.637	.834	.561	.157	-.176
.025	.339	-.112	-1.009	-1.292	-.718	.446	-.045	-.981	-1.060	-.383
.075	.169	-.163	-.850	-1.164	-.729	.345	-.052	-.883	-.462	-.367
.150	.060	-.192	-.556	-1.071	-.731	.195	-.113	-.781	-.487	-.391
.250	-.006	-.197	-.380	-.858	-.702	.039	-.210	-.618	-.501	-.407
.350	-.067	-.230	-.366	-.368	-.428					
.450	-.126	-.291	-.312	-.239	-.407					
.550	-.182	-.353	-.430	-.283	-.417					
.650	-.179	-.337	-.391	-.312	-.430	-.308	-.517	-.516	-.418	-.388
.750	-.123	-.262	-.197	-.287	-.440	-.192	-.298	-.241	-.398	-.387
.850	-.024	-.030	-.029	-.238	-.433	-.092	-.124	-.092	-.358	-.364
.900	.011	.028	.030	-.205	-.429	-.053	-.061	-.034	-.335	-.353
	<i>Right side</i>									
.025	-.952	-.052	.384	.598	.744	-.944	.071	.507	.705	.819
.075	-.780	-.145	.175	.378	.552	-.848	-.028	.348	.561	.729
.150	-.477	-.146	.086	.257	.418	-.688	-.082	.225	.434	.625
.250	-.372	-.179	.007	.158	.306					
.350	-.351	-.217	-.056	.074	.216					
.450	-.314	-.269	-.106	.011	.137					
.550	-.419	-.328	-.156	-.052	.062					
.650	-.374	-.326	-.148	-.084	.006	-.502	-.470	-.241	-.160	-.064
.750	-.213	-.262	-.100	-.102	-.061	-.254	-.300	-.158	-.237	-.184
.850	-.003	.006	.015	-.060	-.080	-.079	-.108	-.037	-.175	-.165
.900	.052	.058	.045	-.072	-.135	-.020	-.041	-.005	-.202	-.217



TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(f)  $\alpha = 9.4^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.442	.821	.473	-.681	-1.137	.132	.691	.105	-.939	-1.220
.025	-.004	-.129	-.258	-.794	-2.065	.355	-.095	-.782	-1.563	-.994
.075	.034	-.118	-.318	-.549	-1.040	.173	-.129	-.486	-1.388	-.966
.150	.000	-.141	-.318	-.455	-.607	.089	-.132	-.357	-.699	-.920
.250	.002	-.093	-.228	-.324	-.420	.025	-.134	-.297	-.340	-.858
.350	-.014	-.102	-.205	-.268	-.347	-.020	-.152	-.267	-.324	-.722
.450	-.032	-.111	-.198	-.238	-.342	-.052	-.159	-.242	-.287	-.575
.550	-.032	-.091	-.159	-.192	-.273	-.036	-.118	-.182	-.234	-.462
.650	-.007	-.057	-.113	-.153	-.211	-.016	-.084	-.134	-.178	-.388
.750	.009	-.025	-.063	-.093	-.158	.009	-.029	-.070	-.107	-.317
.850	.048	.037	.004	-.019	-.091	.055	.034	.018	-.024	-.231
.900	.071	.066	.043	.017	-.045	.082	.068	.057	.020	-.171
	<i>Right side</i>									
.025	-.194	-.036	.022	.534	.815	-.681	-.052	.395	.612	.718
.075	-.328	-.127	.038	.349	.545	-.451	-.107	.195	.402	.547
.150	-.307	-.129	.015	.239	.393	-.344	-.116	.093	.271	.407
.250	-.239	-.100	.004	.186	.299	-.271	-.122	.036	.179	.296
.350	-.212	-.102	-.010	.107	.202	-.250	-.129	-.001	.117	.225
.450	-.198	-.111	-.033	.050	.123	-.225	-.134	-.028	.061	.153
.550	-.150	-.093	-.026	.031	.089	-.180	-.113	-.028	.036	.107
.650	-.100	-.050	-.005	.031	.075	-.123	-.075	-.005	.041	.091
.750	-.052	-.009	.022	.045	.070	-.050	-.013	.027	.057	.075
.850	.023	.048	.064	.073	.080	.032	.057	.068	.087	.068
.900	.055	.068	.073	.082	.082	.059	.073	.080	.080	.038
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.025	.650	-.051	-.635	-.851	.169	.666	-.198	-.621	-.443
.025	.378	-.122	-1.044	-.992	-.632	.506	-.032	-.920	-.773	-.443
.075	.207	-.127	-.578	-.948	-.628	.424	.032	-.684	-.549	-.453
.150	.112	-.132	-.408	-.895	-.616	.337	.021	-.440	-.542	-.480
.250	.062	-.120	-.316	-.780	-.582	.246	-.020	-.272	-.517	-.483
.350	.030	-.125	-.279	-.616	-.561					
.450	.009	-.129	-.251	-.459	-.538					
.550	-.016	-.125	-.215	-.351	-.506					
.650	.005	-.079	-.157	-.257	-.478	-.004	-.109	-.187	-.289	-.407
.750	.025	-.034	-.084	-.181	-.450	-.014	-.079	-.129	-.243	-.386
.850	.046	.014	-.015	-.114	-.402	.000	-.032	-.065	-.197	-.354
.900	.064	.041	.018	-.077	-.388	.021	.003	-.015	-.171	-.328
	<i>Right side</i>									
.025	-.922	-.036	.425	.605	.670	-.925	.093	.556	.653	.690
.075	-.503	-.102	.222	.418	.541	-.581	.071	.441	.614	.732
.150	-.348	-.100	.133	.303	.428	-.314	.071	.374	.568	.723
.250	-.296	-.109	.075	.223	.345					
.350	-.269	-.111	.041	.165	.269					
.450	-.230	-.109	.018	.128	.216					
.550	-.189	-.100	.015	.091	.163					
.650	-.134	-.063	.025	.080	.119	-.164	-.072	.027	.087	.119
.750	-.071	-.009	.036	.066	.068	-.116	-.066	-.005	.011	-.010
.850	.018	.046	.080	.077	.017	-.023	.005	.036	-.003	-.065
.900	.052	.068	.091	.075	-.036	.014	.030	.048	-.031	-.112

TABLE V. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(i)  $\alpha = 9.7^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.344	.925	.460	-.160	.536	.764	.575	.147		
.025	-.081	-.088	-.039	-.455	.335	-.092	-1.001	-1.342		
.075	.036	-.106	-.257	-.488	.163	-.141	-.667	-1.196		
.150	.014	-.159	-.410	-.577	.084	-.143	-.463	-.783		
.250	.001	-.117	-.301	-.481	.018	-.159	-.332	-.519		
.350	-.030	-.127	-.262	-.338	-.042	-.198	-.348	-.424		
.450	-.057	-.162	-.274	-.293	-.088	-.224	-.323	-.396		
.550	-.048	-.127	-.199	-.243	-.072	-.183	-.250	-.301		
.650	-.042	-.100	-.155	-.203	-.052	-.142	-.178	-.246		
.750	-.013	-.055	-.090	-.158	-.013	-.067	-.083	-.172		
.850	.040	.024	-.004	-.073	.052	.029	.020	-.081		
.900	.077	.067	.047	-.019	.092	.078	.075	-.026		
<i>Right side</i>										
.025	.045	.021	-.025	.336	-.908	-.023	.397	.608		
.075	-.262	-.107	.053	.315	-.563	-.099	.197	.390		
.150	-.376	-.127	.046	.252	-.395	-.117	.105	.268		
.250	-.274	-.103	.014	.187	-.309	-.135	.038	.174		
.350	-.262	-.117	-.015	.106	-.325	-.162	-.012	.105		
.450	-.269	-.151	-.047	.027	-.288	-.177	-.051	.041		
.550	-.201	-.126	-.046	.004	-.244	-.161	-.052	.014		
.650	-.129	-.091	-.031	-.009	-.164	-.119	-.028	.012		
.750	-.068	-.043	.004	.012	-.060	-.034	.014	.024		
.850	.021	.039	.062	.047	.040	.052	.074	.044		
.900	.060	.066	.077	.045	.080	.070	.080	.035		
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.433	.711	.450	.027	.453	.743	.272	-.265		
.025	.356	-.138	-1.143	-.875	.511	-.016	-1.023	-1.095		
.075	.195	-.151	-.970	-.830	.446	.012	-.892	-.461		
.150	.101	-.165	-.464	-.797	.340	-.017	-.759	-.450		
.250	.049	-.158	-.387	-.702	.211	-.092	-.588	-.459		
.350	.013	-.173	-.295	-.554						
.450	-.021	-.202	-.348	-.447						
.550	-.057	-.212	-.312	-.374						
.650	-.045	-.165	-.203	-.320	-.081	-.208	-.233	-.404		
.750	-.010	-.082	-.104	-.269	-.062	-.129	-.141	-.370		
.850	.032	.000	-.009	-.220	-.018	-.043	-.056	-.308		
.900	.060	.041	.039	-.187	.009	.001	-.009	-.270		
<i>Right side</i>										
.025	-1.092	-.020	.416	.608	-1.171	.134	.573	.731		
.075	-.868	-.108	.219	.410	-.774	.083	.473	.666		
.150	-.348	-.106	.140	.305	-.508	.062	.390	.591		
.250	-.374	-.127	.077	.222						
.350	-.284	-.143	.042	.164						
.450	-.319	-.165	.008	.114						
.550	-.283	-.171	-.016	.067						
.650	-.209	-.138	-.015	.040	-.234	-.163	-.035	.010		
.750	-.099	-.063	.008	.014	-.139	-.119	-.051	-.081		
.850	.022	.043	.070	.013	-.023	-.004	.026	-.071		
.900	.071	.066	.085	-.013	.022	.028	.040	-.107		

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(j)  $\alpha = 9.7^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.344	.945	.474	-.130		.565	.757	.596	.180	
.025	-.067	-.047	.024	-.390		.339	-.049	-.919	-1.329	
.075	.051	-.064	-.183	-.437		.175	-.103	-.466	-1.194	
.150	.035	-.123	-.344	-.543		.086	-.113	-.404	-.777	
.250	.018	-.095	-.269	-.503		.023	-.140	-.318	-.507	
.350	-.020	-.111	-.249	-.377		-.041	-.186	-.348	-.366	
.450	-.057	-.157	-.278	-.289		-.095	-.222	-.352	-.397	
.550	-.054	-.132	-.211	-.240		-.080	-.187	-.254	-.310	
.650	-.049	-.112	-.159	-.208		-.066	-.148	-.193	-.256	
.750	-.025	-.066	-.091	-.161		-.027	-.076	-.093	-.189	
.850	.029	.018	-.012	-.085		.041	.023	.013	-.097	
.900	.065	.056	.037	-.034		.080	.072	.069	-.044	
	<i>Right side</i>									
.025	.123	.064	.000	.257		-.824	-.002	.408	.617	
.075	-.185	-.064	.072	.312		-.468	-.071	.208	.400	
.150	-.318	-.092	.069	.272		-.355	-.088	.115	.272	
.250	-.246	-.078	.042	.203		-.291	-.113	.045	.176	
.350	-.247	-.103	.003	.113		-.324	-.152	-.007	.100	
.450	-.271	-.145	-.040	.031		-.303	-.176	-.052	.037	
.550	-.206	-.125	-.048	-.001		-.252	-.161	-.058	.004	
.650	-.140	-.098	-.040	-.015		-.178	-.124	-.034	.001	
.750	-.075	-.055	-.005	.004		-.075	-.057	.004	.009	
.850	.012	.035	.050	.032		.029	.051	.065	.031	
.900	.048	.060	.062	.032		.069	.068	.072	.015	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.450	.709	.460	.048		.479	.738	.307	-.225	
.025	.352	-.103	-1.062	-.932		.498	.020	-.903	-1.065	
.075	.193	-.121	-.901	-.851		.430	.031	-.841	-.455	
.150	.097	-.141	-.396	-.820		.326	-.004	-.706	-.446	
.250	.044	-.143	-.383	-.716		.195	-.080	-.559	-.450	
.350	.006	-.165	-.351	-.552						
.450	-.033	-.197	-.332	-.439						
.550	-.070	-.210	-.358	-.375						
.650	-.058	-.176	-.215	-.333		-.107	-.223	-.241	-.418	
.750	-.026	-.092	-.111	-.289		-.080	-.135	-.151	-.387	
.850	.019	-.006	-.016	-.244		-.030	-.047	-.052	-.339	
.900	.047	.039	.032	-.218		.003	-.002	-.009	-.309	
	<i>Right side</i>									
.025	-1.026	-.008	.416	.601		-1.113	.135	.568	.730	
.075	-.726	-.094	.221	.408		-.730	.088	.466	.659	
.150	-.348	-.092	.142	.301		-.451	.060	.381	.585	
.250	-.367	-.116	.077	.221						
.350	-.293	-.135	.037	.163						
.450	-.318	-.164	.001	.109						
.550	-.297	-.177	-.024	.059						
.650	-.235	-.148	-.028	.028		-.254	-.178	-.049	-.007	
.750	-.112	-.092	-.007	-.009		-.150	-.125	-.066	-.105	
.850	.015	.048	.061	-.007		-.031	-.010	.016	-.093	
.900	.063	.066	.076	-.034		.015	.023	.032	-.134	

TABLE V. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(k)  $\alpha = 15.6^\circ$ ;  $M = 0.60$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	-.373	.829	.223	.093	-.424	.299	.730	.282	-.475	-.932
.025	-.606	-.116	.315	.114	-.433	.124	-.100	-.328	-.698	-.951
.075	-.158	-.129	.029	-.083	-.389	.054	-.141	-.303	-.517	-.816
.150	-.040	-.155	-.130	-.191	-.382	.050	-.136	-.264	-.407	-.663
.250	.017	-.118	-.139	-.179	-.324	.040	-.141	-.250	-.345	-.545
.350	.033	-.095	-.160	-.189	-.287	.020	-.155	-.241	-.303	-.449
.450	.038	-.113	-.171	-.186	-.252	.010	-.166	-.220	-.262	-.377
.550	.038	-.097	-.139	-.150	-.201	.022	-.132	-.178	-.216	-.317
.650	.045	-.067	-.095	-.111	-.152	.029	-.102	-.135	-.170	-.261
.750	.047	-.033	-.059	-.074	-.115	.038	-.049	-.077	-.106	-.192
.850	.057	.018	-.010	-.021	-.062	.057	.015	-.012	-.051	-.129
.900	.061	.045	.013	.006	-.025	.070	.041	.022	-.012	-.087
	<i>Right side</i>									
.025	.341	-.113	-.759	-.271	-.097	-.385	-.088	.112	.169	.168
.075	.001	-.175	-.234	-.072	.029	-.311	-.127	.052	.121	.223
.150	-.131	-.150	-.054	.006	.100	-.274	-.129	.043	.114	.254
.250	-.151	-.113	.020	.059	.156	-.246	-.127	.052	.119	.272
.350	-.167	-.095	.038	.080	.182	-.232	-.136	.045	.116	.268
.450	-.179	-.118	.038	.089	.179	-.214	-.141	.029	.110	.251
.550	-.142	-.100	.043	.087	.154	-.172	-.127	.029	.107	.223
.650	-.094	-.061	.045	.082	.128	-.131	-.086	.043	.103	.191
.750	-.057	-.024	.050	.084	.110	-.068	-.040	.052	.098	.161
.850	.001	.025	.066	.084	.093	-.003	.029	.073	.098	.133
.900	.022	.045	.066	.082	.080	.024	.038	.068	.089	.112
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	-.047	.634	.008	-.631	-.781	-.015	.622	-.402	-.838	-.507
.025	.421	-.113	-1.063	-1.269	-.781	.555	.004	-.934	-.686	-.491
.075	.253	-.127	-.582	-1.147	-.753	.498	.066	-.810	-.606	-.454
.150	.163	-.127	-.407	-.838	-.714	.438	.064	-.508	-.574	-.463
.250	.121	-.116	-.310	-.507	-.654	.338	.029	-.314	-.523	-.449
.350	.089	-.120	-.273	-.368	-.577					
.450	.068	-.123	-.245	-.285	-.496					
.550	.050	-.118	-.204	-.239	-.415					
.650	.047	-.081	-.144	-.166	-.331	.045	-.109	-.174	-.205	-.264
.750	.061	-.035	-.079	-.115	-.280	.024	-.081	-.123	-.157	-.217
.850	.063	.004	-.017	-.053	-.220	.029	-.028	-.052	-.106	-.187
.900	.070	.027	.015	-.028	-.194	.043	-.005	-.015	-.083	-.166
	<i>Right side</i>									
.025	-.990	-.049	.451	.591	.571	-1.068	.082	.568	.649	.655
.075	-.562	-.109	.259	.403	.476	-.669	.110	.499	.635	.722
.150	-.366	-.095	.176	.300	.407	-.357	.121	.469	.621	.718
.250	-.299	-.102	.130	.243	.360					
.350	-.267	-.107	.100	.197	.326					
.450	-.228	-.102	.077	.174	.295					
.550	-.191	-.090	.070	.146	.265					
.650	-.133	-.063	.064	.132	.216	-.131	-.054	.084	.142	.212
.750	-.066	-.031	.066	.116	.172	-.100	-.063	.036	.061	.098
.850	.010	.029	.084	.110	.126	-.017	.004	.059	.052	.056
.900	.043	.052	.093	.098	.087	.010	.027	.059	.038	.008

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Continued

(i)  $\alpha = 15.8^\circ$ ;  $M = 0.80$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.005	.876	.477	.096	-.327	.319	.785	.282	-.285	-.579
.025	-.920	.122	.336	.017	-.444	-.012	-.060	-.326	-.554	-.654
.075	-.513	-.079	.022	-.136	-.400	-.070	-.148	-.326	-.468	-.601
.150	-.241	-.183	-.185	-.241	-.384	-.053	-.152	-.302	-.407	-.525
.250	-.119	-.141	-.178	-.212	-.333	-.030	-.155	-.282	-.366	-.472
.350	-.050	-.100	-.187	-.203	-.283	-.026	-.169	-.276	-.336	-.420
.450	-.009	-.120	-.204	-.195	-.238	-.018	-.178	-.248	-.288	-.357
.550	.016	-.088	-.145	-.140	-.184	.008	-.138	-.202	-.238	-.303
.650	.034	-.047	-.095	-.093	-.141	.020	-.099	-.149	-.185	-.257
.750	.046	-.016	-.058	-.067	-.105	.040	-.047	-.087	-.121	-.195
.850	.060	.030	-.012	-.017	-.049	.063	.021	-.018	-.055	-.137
.900	.068	.060	.014	.009	-.010	.077	.060	.023	-.015	-.081
<i>Right side</i>										
.025	.374	-.765	-.952	-.419	-.294	-.408	-.232	-.043	-.014	.026
.075	-.022	-.449	-.596	-.238	-.171	-.355	-.242	-.084	-.040	.094
.150	-.189	-.238	-.277	-.157	-.050	-.318	-.193	-.064	-.015	.148
.250	-.189	-.134	-.124	-.075	.053	-.284	-.144	-.026	.021	.214
.350	-.200	-.063	-.055	-.014	.117	-.266	-.125	-.009	.045	.233
.450	-.209	-.062	-.012	.029	.154	-.240	-.125	-.001	.064	.242
.550	-.151	-.042	.017	.052	.154	-.197	-.097	.014	.079	.232
.650	-.090	-.015	.034	.067	.139	-.137	-.056	.034	.091	.212
.750	-.048	.015	.048	.070	.148	-.067	-.007	.052	.102	.185
.850	-.001	.056	.068	.087	.106	-.003	.051	.075	.102	.156
.900	.017	.067	.068	.084	.094	.027	.068	.075	.097	.132
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.339	.680	.425	.023	-.375	.278	.684	.057	-.468	-.439
.025	.350	-.088	-1.104	-1.150	-.640	.556	.022	-.871	-.772	-.445
.075	.189	-.126	-.580	-1.006	-.618	.498	.080	-.739	-.601	-.391
.150	.114	-.135	-.422	-.784	-.588	.419	.056	-.615	-.569	-.390
.250	.089	-.125	-.320	-.520	-.545	.305	.008	-.342	-.544	-.381
.350	.074	-.126	-.282	-.383	-.503					
.450	.051	-.137	-.259	-.304	-.442					
.550	.034	-.131	-.216	-.246	-.381					
.650	.039	-.091	-.149	-.169	-.309	.014	-.129	-.188	-.217	-.238
.750	.056	-.044	-.080	-.113	-.266	-.001	-.094	-.124	-.159	-.208
.850	.066	.010	-.003	-.049	-.220	.016	-.028	-.047	-.102	-.168
.900	.077	.042	.028	-.020	-.192	.034	.011	-.006	-.076	-.145
<i>Right side</i>										
.025	-1.185	-.091	.374	.562	.641	-.897	.100	.565	.696	.774
.075	-.562	-.129	.181	.358	.519	-.673	.128	.496	.666	.808
.150	-.387	-.091	.123	.268	.451	-.568	.128	.445	.617	.779
.250	-.324	-.086	.095	.218	.400					
.350	-.287	-.089	.072	.183	.367					
.450	-.249	-.085	.060	.161	.340					
.550	-.206	-.076	.046	.138	.303					
.650	-.145	-.050	.049	.123	.260	-.142	-.053	.048	.123	.248
.750	-.071	-.008	.059	.113	.211	-.105	-.060	.011	.052	.127
.850	.020	.056	.092	.119	.159	-.022	.019	.049	.052	.102
.900	.057	.077	.097	.102	.112	.017	.042	.059	.029	.029

TABLE V.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE REARWARD POSITION;  $i_t = 0^\circ$  - Concluded

(m)  $\alpha = 15.9^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.312	.905	.652	.075		.372	.781	.290	-.311	
.025	-1.074	.049	.196	-.117		-.016	-.143	-.198	-.632	
.075	-.562	-.157	-.073	-.246		-.126	-.226	-.456	-.543	
.150	-.315	-.276	-.297	-.345		-.133	-.220	-.416	-.484	
.250	-.209	-.219	-.271	-.287		-.116	-.204	-.368	-.427	
.350	-.145	-.159	-.247	-.260		-.110	-.217	-.338	-.380	
.450	-.094	-.169	-.256	-.237		-.093	-.230	-.310	-.337	
.550	-.047	-.119	-.181	-.172		-.050	-.177	-.253	-.274	
.650	-.013	-.075	-.130	-.123		-.024	-.133	-.194	-.219	
.750	.013	-.035	-.087	-.098		.003	-.075	-.130	-.150	
.850	.037	.021	-.034	-.038		.040	.002	-.053	-.078	
.900	.050	.046	-.007	-.011		.057	.049	-.013	-.033	
	<i>Right side</i>									
.025	.282	-.694	-1.007	-.489		-.618	-.280	-.081	-.096	
.075	-.100	-.882	-.668	-.330		-.504	-.308	-.168	-.126	
.150	-.312	-.323	-.341	-.246		-.431	-.254	-.163	-.106	
.250	-.279	-.229	-.238	-.169		-.352	-.199	-.131	-.059	
.350	-.258	-.149	-.167	-.102		-.318	-.179	-.101	-.033	
.450	-.250	-.129	-.110	-.046		-.286	-.173	-.078	-.004	
.550	-.179	-.085	-.058	-.006		-.233	-.142	-.051	.024	
.650	-.112	-.043	-.021	.022		-.172	-.090	-.018	.049	
.750	-.066	-.012	.003	.014		-.093	-.035	.014	.071	
.850	-.007	.041	.042	.063		-.020	.035	.049	.086	
.900	.009	.052	.047	.062		.013	.051	.049	.075	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.378	.669	.460	.160		.315	.689	.154	-.350	
.025	.299	-.139	-1.228	-1.174		.528	.005	-1.140	-.833	
.075	.134	-.176	-.737	-1.009		.475	.056	-.766	-.615	
.150	.059	-.183	-.497	-.794		.386	.022	-.633	-.566	
.250	.034	-.166	-.363	-.573		.257	-.035	-.436	-.558	
.350	.020	-.166	-.328	-.445						
.450	-.003	-.177	-.308	-.360						
.550	-.019	-.170	-.263	-.290						
.650	-.009	-.132	-.186	-.214		-.037	-.179	-.218	-.264	
.750	.014	-.080	-.107	-.147		-.046	-.137	-.153	-.202	
.850	.037	-.012	-.021	-.090		-.016	-.058	-.071	-.145	
.900	.051	.025	.013	-.051		.007	-.012	-.027	-.170	
	<i>Right side</i>									
.025	-1.235	-.132	.309	.529		-1.104	.072	.539	.697	
.075	-.761	-.183	.112	.313		-.705	.096	.469	.659	
.150	-.422	-.146	.057	.226		-.614	.093	.410	.605	
.250	-.352	-.133	.036	.180						
.350	-.331	-.126	.014	.146						
.450	-.285	-.126	.006	.126						
.550	-.243	-.119	-.010	.108						
.650	-.173	-.092	.002	.096		-.172	-.105	.009	.092	
.750	-.092	-.045	.017	.083		-.132	-.100	-.036	.014	
.850	.009	.038	.062	.091		-.037	-.012	.019	.025	
.900	.049	.063	.069	.078		.003	.026	.026	-.006	

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $\alpha_1 = 0^\circ$

(-)  $\alpha = 0^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.607	.665	.360	-.480	-1.371	.306	.748	.335	-.433	-1.234
.025	.272	-.098	-.532	-1.152	-2.378	.332	-.091	-.624	-1.087	-1.186
.075	.131	-.109	-.371	-.695	-1.544	.144	-.128	-.428	-.960	-1.154
.150	.038	-.137	-.297	-.487	-.671	.052	-.139	-.318	-.714	-1.105
.250	.022	-.095	-.201	-.308	-.433	-.013	-.146	-.267	-.442	-1.043
.350	-.032	-.112	-.175	-.245	-.348	-.064	-.160	-.249	-.343	-.876
.450	-.064	-.123	-.162	-.208	-.301	-.103	-.165	-.231	-.285	-.659
.550	-.055	-.102	-.127	-.169	-.318	-.099	-.144	-.175	-.222	-.498
.650	-.038	-.077	-.086	-.127	-.248	-.078	-.107	-.120	-.159	-.375
.750	-.022	-.036	-.040	-.092	-.170	-.038	-.047	-.061	-.094	-.262
.850	.019	.011	.008	-.034	-.082	.019	.020	.022	-.009	-.144
.900	.040	.045	.045	.010	-.034	.050	.057	.052	.033	-.087
	<i>Right side</i>									
.025	-.460	-.029	.328	.606	.830	-.631	-.063	.379	.604	.740
.075	-.381	-.109	.130	.353	.551	-.427	-.118	.160	.363	.535
.150	-.302	-.125	.052	.212	.378	-.325	-.132	.064	.223	.375
.250	-.207	-.100	.020	.138	.271	-.261	-.139	-.003	.121	.244
.350	-.180	-.107	-.019	.070	.172	-.233	-.146	-.049	.052	.156
.450	-.161	-.125	-.061	.017	.105	-.182	-.135	-.079	-.025	.043
.550	-.119	-.093	-.040	.019	.087	-.175	-.132	-.079	-.018	.045
.650	-.080	-.059	-.021	.026	.077	-.113	-.091	-.047	-.004	.036
.750	-.034	-.022	.004	.045	.077	-.050	-.033	-.010	.024	.036
.850	.029	.031	.050	.075	.091	.033	.043	.054	.066	.054
.900	.054	.059	.059	.073	.087	.063	.061	.061	.061	.043
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.216	.720	.264	-.429	-.701	.408	.681	.231	-.329	-.505
.025	.350	-.123	-.824	-1.092	-.759	.214	-.211	-.736	-.851	-.537
.075	.154	-.144	-.486	-.990	-.740	.031	-.162	-.382	-.837	-.549
.150	.040	-.158	-.364	-.804	-.726	-.057	-.151	-.284	-.726	-.604
.250	-.032	-.165	-.286	-.552	-.694	-.103	-.162	-.251	-.540	-.599
.350	-.085	-.174	-.254	-.377	-.659					
.450	-.117	-.176	-.226	-.292	-.625					
.550	-.108	-.151	-.187	-.229	-.585					
.650	-.085	-.116	-.125	-.166	-.528	-.103	-.091	-.120	-.347	-.410
.750	-.050	-.047	-.056	-.094	-.465	-.071	-.038	-.079	-.329	-.382
.850	.008	.008	.002	-.032	-.387	-.029	-.003	-.061	-.299	-.357
.900	.029	.029	.027	-.004	-.359	-.008	.017	-.049	-.264	-.341
	<i>Right side</i>									
.025	-.726	-.072	.392	.597	.701	-.640	-.135	.268	.423	.498
.075	-.483	-.135	.151	.358	.491	-.367	-.165	.031	.142	.246
.150	-.339	-.137	.066	.216	.343	-.279	-.148	-.047	.005	.096
.250	-.284	-.155	.012	.098	.211					
.350	-.258	-.169	-.077	.022	.112					
.450	-.221	-.162	-.100	-.032	.040					
.550	-.175	-.142	-.102	-.053	.003					
.650	-.117	-.100	-.067	-.041	-.015	.031	-.001	-.042	-.055	-.080
.750	-.052	-.042	-.031	-.006	-.027	-.080	-.033	-.054	-.074	-.121
.850	.029	.036	.041	.047	-.024	-.022	.015	-.003	-.032	-.124
.900	.056	.054	.061	.052	-.057	-.008	.031	.011	-.027	-.147

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE L/W POSITION;  $i_t = 0^\circ$  - Continued

(b)  $\alpha = 0^\circ$ ;  $M = 0.80$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.717	.677	.545	.183	-.324	.490	.781	.545	.045	-.586
.025	.281	-.055	-.527	-1.170	-1.565	.335	-.070	-.955	-1.686	-1.006
.075	.146	-.088	-.371	-.909	-1.284	.152	-.117	-.449	-1.420	-1.022
.150	.043	-.123	-.324	-.529	-1.007	.055	-.132	-.344	-.597	-.961
.250	.017	-.090	-.214	-.306	-.623	-.012	-.147	-.278	-.400	-.877
.350	-.024	-.109	-.187	-.251	-.399	-.075	-.168	-.268	-.344	-.774
.450	-.072	-.126	-.173	-.223	-.329	-.121	-.183	-.246	-.292	-.656
.550	-.063	-.102	-.133	-.175	-.304	-.109	-.151	-.185	-.219	-.543
.650	-.047	-.072	-.092	-.141	-.232	-.084	-.108	-.123	-.149	-.445
.750	-.027	-.037	-.043	-.098	-.188	-.035	-.036	-.054	-.069	-.352
.850	.016	.026	.022	-.028	-.119	.028	.044	.035	.019	-.249
.900	.042	.068	.058	.016	-.067	.068	.088	.071	.059	-.189
	<i>Right side</i>									
.025	-.415	.017	.338	.601	.827	-.958	-.031	.377	.604	.751
.075	-.392	-.084	.140	.354	.566	-.429	-.100	.163	.364	.541
.150	-.310	-.109	.059	.221	.389	-.331	-.117	.065	.227	.388
.250	-.210	-.085	.024	.149	.287	-.268	-.133	-.004	.126	.262
.350	-.187	-.108	-.025	.071	.184	-.245	-.153	-.051	.050	.174
.450	-.167	-.120	-.065	.015	.105	-.190	-.138	-.089	-.020	.056
.550	-.118	-.085	-.048	.015	.086	-.176	-.138	-.098	-.031	.034
.650	-.070	-.046	-.022	.027	.074	-.106	-.088	-.063	-.014	.022
.750	-.024	-.010	-.002	.036	.063	-.026	-.008	-.011	.018	.017
.850	.037	.050	.042	.065	.074	.059	.068	.059	.068	.027
.900	.068	.077	.058	.073	.060	.083	.089	.077	.077	.008
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.427	.751	.477	.004	-.431	.502	.710	.390	.015	-.333
.025	.345	-.108	-1.091	-1.194	-.702	.209	-.228	-1.010	-.707	-.476
.075	.161	-.142	-.567	-1.120	-.702	.008	-.183	-.581	-.701	-.490
.150	.040	-.163	-.399	-.959	-.699	-.076	-.159	-.294	-.701	-.542
.250	-.040	-.171	-.312	-.778	-.669	-.124	-.168	-.248	-.617	-.546
.350	-.099	-.184	-.271	-.581	-.630					
.450	-.133	-.189	-.239	-.399	-.588					
.550	-.132	-.162	-.193	-.262	-.543					
.650	-.095	-.109	-.121	-.156	-.499	-.093	-.067	-.118	-.358	-.388
.750	-.038	-.019	-.045	-.072	-.448	-.053	-.007	-.069	-.310	-.369
.850	.019	.038	.024	-.008	-.398	-.003	.038	-.046	-.274	-.339
.900	.042	.067	.051	.021	-.373	.011	.061	-.030	-.237	-.324
	<i>Right side</i>									
.025	-1.060	-.040	.389	.598	.719	-.969	-.135	.254	.419	.530
.075	-.503	-.120	.164	.360	.518	-.461	-.180	-.004	.128	.267
.150	-.353	-.123	.064	.222	.365	-.271	-.154	-.080	-.008	.097
.250	-.302	-.151	-.033	.102	.226					
.350	-.268	-.178	-.091	.010	.115					
.450	-.227	-.171	-.124	-.045	.037					
.550	-.178	-.141	-.121	-.065	-.016					
.650	-.106	-.088	-.080	-.048	-.036	.013	.007	-.040	-.055	-.097
.750	-.033	-.020	-.033	-.017	-.053	-.055	-.008	-.051	-.068	-.133
.850	.054	.070	.058	.050	-.041	.005	.061	.016	-.020	-.122
.900	.091	.100	.083	.064	-.079	.016	.076	.029	-.023	-.145



TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(c)  $\alpha = 0^\circ$ ;  $M = 0.85$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.742	.680	.587	.301	-.144	.537	.787	.600	.187	-.403
.025	.280	-.059	-.528	-1.095	-1.341	.326	-.079	-1.088	-1.559	-.914
.075	.142	-.095	-.384	-.874	-1.153	.144	-.136	-.426	-1.395	-.937
.150	.043	-.145	-.351	-.670	-.973	.050	-.155	-.356	-1.209	-.930
.250	.017	-.109	-.228	-.336	-.716	-.026	-.169	-.304	-.336	-.858
.350	-.036	-.131	-.198	-.255	-.516	-.090	-.194	-.290	-.293	-.757
.450	-.090	-.152	-.189	-.211	-.407	-.146	-.209	-.267	-.271	-.648
.550	-.074	-.122	-.142	-.159	-.338	-.131	-.179	-.192	-.211	-.556
.650	-.063	-.089	-.101	-.134	-.256	-.106	-.132	-.131	-.142	-.470
.750	-.040	-.051	-.046	-.088	-.216	-.046	-.048	-.053	-.064	-.380
.850	.007	.012	.018	-.015	-.134	.030	.032	.034	.029	-.276
.900	.039	.055	.058	.030	-.076	.067	.076	.076	.069	-.217
	<i>Right side</i>									
.025	-.388	.012	.338	.608	.833	-.986	-.049	.371	.598	.750
.075	-.397	-.097	.141	.359	.563	-.436	-.124	.159	.360	.543
.150	-.328	-.129	.054	.223	.399	-.354	-.139	.058	.227	.386
.250	-.226	-.108	.015	.155	.290	-.288	-.156	-.012	.125	.272
.350	-.200	-.128	-.040	.072	.187	-.268	-.175	-.069	.046	.169
.450	-.184	-.142	-.078	.006	.103	-.204	-.165	-.103	-.025	.056
.550	-.127	-.106	-.066	.003	.079	-.184	-.166	-.116	-.042	.029
.650	-.080	-.064	-.038	.025	.072	-.116	-.108	-.079	-.029	.007
.750	-.027	-.029	-.019	.033	.054	-.030	-.027	-.019	.009	.009
.850	.037	.039	.038	.068	.064	.056	.062	.054	.069	.014
.900	.070	.062	.056	.068	.050	.089	.079	.078	.079	-.016
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.470	.757	.533	.125	-.331	.522	.711	.435	.010	-.306
.025	.336	-.114	-1.210	-1.641	-.746	.187	-.262	-1.089	-.975	-.504
.075	.154	-.158	-.579	-1.473	-.758	-.024	-.218	-.678	-.907	-.530
.150	.033	-.188	-.420	-1.291	-.758	-.106	-.188	-.310	-.865	-.593
.250	-.057	-.198	-.321	-.500	-.700	-.141	-.201	-.261	-.706	-.581
.350	-.121	-.216	-.278	-.355	-.666					
.450	-.161	-.221	-.254	-.285	-.620					
.550	-.154	-.186	-.199	-.199	-.570					
.650	-.108	-.132	-.128	-.122	-.514	-.108	-.082	-.128	-.396	-.414
.750	-.043	-.028	-.045	-.051	-.450	-.050	-.022	-.078	-.350	-.393
.850	.020	.028	.030	.020	-.387	-.006	.029	-.053	-.301	-.366
.900	.047	.060	.058	.042	-.366	.013	.053	-.032	-.262	-.350
	<i>Right side</i>									
.025	-1.163	-.058	.382	.588	.717	-1.138	-.164	.245	.423	.530
.075	-.436	-.144	.154	.360	.516	-.490	-.218	-.032	.119	.267
.150	-.371	-.148	.057	.217	.364	-.270	-.189	-.106	-.041	.086
.250	-.314	-.179	-.043	.095	.226					
.350	-.281	-.213	-.108	.000	.107					
.450	-.241	-.204	-.151	-.062	.023					
.550	-.186	-.174	-.138	-.084	-.034					
.650	-.111	-.105	-.093	-.055	-.051	-.038	-.049	-.055	-.075	-.109
.750	-.034	-.032	-.042	-.025	-.064	-.060	-.019	-.049	-.068	-.153
.850	.059	.059	.060	.055	-.053	.003	.050	.018	-.018	-.133
.900	.096	.092	.081	.072	-.084	.020	.072	.034	-.018	-.158

TABLE VI. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(d)  $\alpha = 0^\circ$ ;  $M = 0.90$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>					<i>Right side</i>				
.000	.773	.689	.639	.409	.008	.595	.801	.661	.321	-.228
.025	.291	-.048	-.468	-.993	-1.187	.328	-.075	-.993	-1.337	-.802
.075	.154	-.089	-.377	-.782	-1.027	.154	-.137	-.556	-1.213	-.815
.150	.045	-.141	-.355	-.660	-.882	.052	-.157	-.359	-1.104	-.833
.250	.024	-.109	-.227	-.489	-.691	-.022	-.172	-.307	-.620	-.799
.350	-.034	-.136	-.206	-.327	-.557	-.096	-.210	-.325	-.431	-.722
.450	-.099	-.163	-.196	-.262	-.474	-.160	-.231	-.264	-.268	-.636
.550	-.084	-.128	-.136	-.165	-.390	-.154	-.195	-.188	-.154	-.549
.650	-.073	-.097	-.089	-.124	-.317	-.124	-.141	-.119	-.106	-.479
.750	-.049	-.052	-.033	-.078	-.272	-.058	-.043	-.039	-.042	-.408
.850	.005	.019	.039	-.006	-.178	.032	.043	.050	.041	-.324
.900	.038	.058	.071	.034	-.114	.071	.086	.092	.083	-.265
	<i>Left side</i>					<i>Right side</i>				
.025	-.339	.030	.348	.605	.837	-.935	-.036	.368	.588	.756
.075	-.411	-.085	.158	.362	.572	-.606	-.117	.166	.357	.542
.150	-.356	-.120	.062	.228	.407	-.375	-.140	.062	.226	.390
.250	-.240	-.105	.024	.151	.296	-.304	-.163	-.010	.123	.268
.350	-.214	-.129	-.036	.067	.190	-.296	-.194	-.071	.035	.171
.450	-.197	-.155	-.085	-.005	.099	-.215	-.182	-.117	-.044	.060
.550	-.125	-.112	-.068	-.005	.079	-.186	-.172	-.128	-.064	.017
.650	-.073	-.070	-.036	.013	.070	-.109	-.113	-.089	-.048	-.001
.750	-.022	-.031	-.016	.021	.045	-.019	-.021	-.020	-.002	-.008
.850	.048	.043	.039	.059	.043	.069	.065	.060	.059	-.005
.900	.076	.069	.058	.067	.020	.100	.092	.082	.078	-.035
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>					<i>Right side</i>				
.000	.526	.769	.597	.253	-.197	.556	.722	.495	.142	-.230
.025	.335	-.113	-1.116	-1.417	-.750	.186	-.287	-1.079	-1.333	-.521
.075	.150	-.163	-.908	-1.278	-.764	-.042	-.260	-1.047	-1.341	-.536
.150	.032	-.199	-.383	-1.202	-.766	-.170	-.210	-.343	-1.084	-.589
.250	-.065	-.219	-.393	-1.112	-.671	-.162	-.225	-.191	-.710	-.580
.350	-.140	-.249	-.239	-.666	-.640					
.450	-.189	-.246	-.239	-.162	-.613					
.550	-.181	-.206	-.195	-.098	-.580					
.650	-.127	-.137	-.115	-.065	-.538	-.113	-.078	-.117	-.323	-.439
.750	-.060	-.025	-.029	-.017	-.481	-.032	-.011	-.055	-.305	-.414
.850	.033	.034	.042	.038	-.445	.013	.043	-.037	-.277	-.386
.900	.052	.067	.074	.057	-.387	.029	.063	-.021	-.240	-.374
	<i>Left side</i>					<i>Right side</i>				
.025	-1.077	-.048	.377	.576	.717	-1.068	-.167	.238	.410	.535
.075	-.857	-.147	.158	.342	.517	-.990	-.262	-.060	.099	.273
.150	-.346	-.152	.054	.209	.369	-.245	-.211	-.172	-.085	.082
.250	-.378	-.195	-.047	.079	.229					
.350	-.244	-.233	-.127	-.019	.112					
.450	-.237	-.227	-.172	-.105	.017					
.550	-.189	-.187	-.158	-.124	-.047					
.650	-.105	-.110	-.103	-.082	-.071	-.060	-.062	-.049	-.077	-.126
.750	-.027	-.025	-.036	-.034	-.084	-.054	-.012	-.043	-.074	-.173
.850	.075	.075	.064	.054	-.062	.010	.063	.036	.011	-.143
.900	.109	.108	.090	.075	-.094	.021	.080	.047	-.009	-.165

TABLE VI. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(e)  $\alpha = 0^\circ$ ;  $M = 0.92$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.778	.692	.656	.446	.066	.621	.799	.677	.365	-.168
.025	.289	-.041	-.449	-.953	-1.170	.322	-.073	-.953	-1.269	-.772
.075	.155	-.085	-.378	-.751	-1.003	.149	-.141	-.683	-1.156	-.781
.150	.047	-.143	-.358	-.639	-.855	.049	-.162	-.411	-1.058	-.802
.250	.020	-.111	-.243	-.542	-.666	-.027	-.183	-.314	-.671	-.775
.350	-.039	-.142	-.218	-.370	-.556	-.105	-.226	-.341	-.508	-.718
.450	-.107	-.180	-.219	-.315	-.498	-.174	-.264	-.331	-.424	-.642
.550	-.093	-.143	-.151	-.203	-.416	-.177	-.219	-.182	-.159	-.570
.650	-.084	-.112	-.097	-.130	-.356	-.140	-.159	-.122	-.081	-.508
.750	-.057	-.061	-.032	-.073	-.311	-.065	-.041	-.036	-.020	-.437
.850	.000	.015	.036	-.001	-.216	.030	.038	.052	.057	-.356
.900	.035	.061	.068	.044	-.150	.075	.087	.096	.096	-.302
<i>Right side</i>										
.025	-.319	.034	.347	.612	.845	-.873	-.036	.363	.590	.759
.075	-.400	-.085	.154	.368	.576	-.635	-.128	.158	.360	.549
.150	-.369	-.125	.062	.233	.410	-.408	-.146	.056	.227	.395
.250	-.255	-.115	.020	.157	.299	-.306	-.170	-.020	.119	.272
.350	-.229	-.141	-.045	.064	.189	-.322	-.210	-.081	.036	.169
.450	-.209	-.171	-.098	-.010	.098	-.258	-.215	-.137	-.051	.060
.550	-.137	-.129	-.082	-.013	.077	-.177	-.200	-.152	-.079	.015
.650	-.072	-.078	-.051	.006	.064	-.102	-.125	-.103	-.060	-.014
.750	-.023	-.036	-.032	.014	.036	-.017	-.022	-.032	-.010	-.021
.850	.050	.041	.036	.058	.027	.072	.066	.052	.061	-.021
.900	.083	.070	.052	.066	-.007	.104	.091	.081	.078	-.058
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.546	.770	.622	.301	-.138	.576	.721	.514	.198	-.200
.025	.326	-.113	-1.077	-1.345	-.726	.173	-.285	-1.047	-1.303	-.519
.075	.148	-.170	-.892	-1.213	-.731	-.054	-.302	-1.023	-1.309	-.537
.150	.025	-.209	-.516	-1.150	-.739	-.205	-.248	-.705	-1.092	-.582
.250	-.071	-.239	-.394	-1.095	-.689	-.224	-.253	-.161	-.964	-.581
.350	-.161	-.294	-.391	-.985	-.654					
.450	-.226	-.279	-.198	-.398	-.609					
.550	-.203	-.234	-.193	-.056	-.579					
.650	-.140	-.142	-.116	-.020	-.550	-.098	-.081	-.119	-.342	-.451
.750	-.051	-.028	-.034	.014	-.495	-.033	-.011	-.064	-.307	-.432
.850	.037	.037	.041	.058	-.436	.008	.041	-.040	-.273	-.405
.900	.054	.072	.074	.079	-.414	.028	.064	-.024	-.236	-.392
<i>Right side</i>										
.025	-1.017	-.054	.367	.574	.720	-.999	-.175	.227	.411	.544
.075	-.830	-.154	.149	.346	.515	-.971	-.320	-.072	.105	.274
.150	-.441	-.164	.045	.215	.372	-.597	-.248	-.204	-.085	.081
.250	-.386	-.219	-.061	.080	.229					
.350	-.390	-.285	-.152	-.028	.107					
.450	-.196	-.257	-.204	-.119	.005					
.550	-.180	-.209	-.182	-.147	-.062					
.650	-.099	-.116	-.120	-.099	-.083	-.091	-.077	-.068	-.074	-.149
.750	-.021	-.030	-.043	-.040	-.102	-.059	-.014	-.048	-.062	-.188
.850	.080	.078	.065	.058	-.074	.009	.063	.031	.000	-.158
.900	.110	.104	.091	.084	-.111	.021	.075	.047	.006	-.175

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $t_t = 0^\circ$  - Continued

(f)  $\alpha = 9.4^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.404	.799	.443	-.682	-1.174	.155	.683	.157	-.727	-1.512
.025	-.027	-.136	-.234	-.727	-2.628	.336	-.108	-.727	-1.225	-1.336
.075	.016	-.133	-.318	-.550	-.863	.150	-.145	-.475	-1.032	-1.292
.150	-.022	-.156	-.313	-.457	-.658	.055	-.154	-.352	-.777	-1.372
.250	-.018	-.113	-.229	-.332	-.429	-.011	-.165	-.291	-.466	-1.192
.350	-.047	-.124	-.209	-.282	-.361	-.067	-.181	-.282	-.347	-.801
.450	-.070	-.142	-.202	-.254	-.361	-.102	-.204	-.259	-.302	-.464
.550	-.079	-.133	-.184	-.222	-.295	-.104	-.165	-.209	-.241	-.333
.650	-.058	-.111	-.150	-.191	-.235	-.083	-.129	-.161	-.191	-.285
.750	-.047	-.079	-.109	-.125	-.178	-.054	-.083	-.093	-.118	-.208
.850	-.013	-.031	-.050	-.072	-.105	.005	-.013	-.018	-.032	-.114
.900	.012	.005	-.007	-.027	-.057	.030	.016	-.018	.009	-.066
	<i>Right side</i>									
.025	.181	-.054	.005	.498	.813	-.698	-.072	.375	.580	.708
.075	.333	-.142	.018	.316	.548	-.464	-.129	.168	.366	.543
.150	.315	-.142	-.004	.212	.388	-.351	-.142	.071	.232	.388
.250	.238	-.106	-.022	.155	.285	-.287	-.154	.005	.132	.269
.350	.224	-.122	-.047	.082	.194	-.269	-.165	-.047	.068	.183
.450	.215	-.138	-.066	.028	.107	-.244	-.183	-.084	.007	.103
.550	.174	-.117	-.057	.012	.080	-.208	-.163	-.088	-.020	.055
.650	.142	-.090	-.041	.012	.071	-.147	-.120	-.066	-.020	.039
.750	.102	-.065	-.022	.012	.050	-.081	-.061	-.032	.003	.036
.850	.031	-.009	.021	.048	.073	-.002	.007	.028	.055	.059
.900	.003	.019	.039	.053	.071	.028	.023	.032	.041	.043
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.032	.626	.009	-.795	-1.123	.182	.570	-.043	-.463	-.854
.025	.336	-.142	-1.022	-1.325	-.909	.209	-.215	-.888	-.727	-.612
.075	.157	-.172	-.543	-1.297	-.895	.007	-.201	-.482	-.709	-.601
.150	.041	-.188	-.402	-1.193	-.863	-.092	-.188	-.309	-.672	-.610
.250	-.033	-.185	-.318	-.802	-.827	-.077	-.126	-.222	-.575	-.598
.350	-.088	-.197	-.284	-.382	-.797					
.450	-.115	-.199	-.254	-.254	-.749					
.550	-.131	-.190	-.213	-.220	-.676					
.650	-.099	-.142	-.152	-.168	-.591	-.115	-.099	-.143	-.368	-.480
.750	-.056	-.088	-.088	-.100	-.502	-.079	-.054	-.111	-.329	-.450
.850	-.018	-.024	-.027	-.043	-.397	-.047	-.015	-.082	-.284	-.429
.900	.003	-.006	-.004	-.016	-.356	-.024	.005	-.063	-.259	-.413
	<i>Right side</i>									
.025	-.931	-.083	.382	.566	.639	-.748	-.149	.246	.371	.445
.075	-.519	-.147	.164	.350	.498	-.437	-.201	-.004	.109	.230
.150	-.360	-.151	.062	.223	.354	-.276	-.172	-.091	-.016	.073
.250	-.312	-.174	-.018	.107	.235					
.350	-.281	-.188	-.079	.025	.132					
.450	-.247	-.190	-.109	-.025	.059					
.550	-.204	-.172	-.116	-.050	.016					
.650	-.147	-.138	-.084	-.043	-.012	-.117	-.088	-.111	-.127	-.149
.750	-.079	-.056	-.043	-.022	-.021	-.092	-.061	-.086	-.118	-.167
.850	.003	.007	.023	.025	-.018	-.036	.005	-.027	-.077	-.167
.900	.028	.025	.043	.039	-.037	-.022	.016	-.013	-.077	-.190

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(g)  $\alpha = 9.6^\circ$ ;  $M = 0.80$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.355	.853	.427	-.364	-.157	.405	.724	.442	-.136	-.807
.025	-.048	-.110	-.142	-.590	-1.622	.324	-.097	-.955	-1.493	-1.070
.075	.017	-.128	-.313	-.566	-1.094	.148	-.142	-.496	-1.149	-1.047
.150	-.016	-.162	-.352	-.533	-.702	.054	-.151	-.370	-.816	-1.006
.250	-.022	-.107	-.246	-.372	-.478	-.013	-.165	-.310	-.469	-.951
.350	-.055	-.122	-.225	-.305	-.397	-.078	-.195	-.299	-.386	-.811
.450	-.078	-.157	-.223	-.275	-.361	-.134	-.216	-.276	-.340	-.639
.550	-.084	-.141	-.195	-.239	-.329	-.125	-.185	-.222	-.275	-.505
.650	-.067	-.118	-.158	-.205	-.289	-.108	-.147	-.167	-.205	-.415
.750	-.054	-.087	-.119	-.154	-.256	-.064	-.081	-.093	-.128	-.339
.850	-.013	-.028	-.049	-.076	-.177	.004	-.001	-.005	-.030	-.233
.900	.017	.011	-.004	-.031	-.120	.040	.042	.042	.017	-.166
	<i>Right side</i>									
.025	-.099	-.019	-.019	.393	.779	-1.080	-.048	.380	.589	.717
.075	-.334	-.130	.031	.311	.530	-.467	-.121	.172	.366	.536
.150	-.346	-.147	.010	.224	.381	-.367	-.130	.077	.240	.394
.250	-.260	-.104	-.011	.161	.280	-.304	-.150	.008	.143	.269
.350	-.242	-.121	-.043	.081	.180	-.287	-.171	-.045	.066	.181
.450	-.231	-.148	-.069	.019	.096	-.264	-.183	-.089	-.004	.091
.550	-.190	-.125	-.057	-.001	.069	-.217	-.168	-.105	-.037	.041
.650	-.148	-.095	-.039	.007	.055	-.155	-.125	-.075	-.031	.021
.750	-.110	-.065	-.022	.007	.032	-.075	-.056	-.028	-.007	.011
.850	-.030	-.001	.024	.042	.044	.013	.028	.039	.042	.027
.900	.004	.025	.042	.049	.034	.039	.048	.051	.045	-.001
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.284	.659	.296	-.252	-.723	.314	.603	.170	-.236	-.659
.025	.331	-.138	-1.183	-1.104	-.729	.193	-.251	-.898	-.667	-.551
.075	.154	-.165	-.642	-1.052	-.723	-.020	-.230	-.684	-.645	-.543
.150	.042	-.180	-.431	-1.007	-.708	-.136	-.195	-.363	-.657	-.546
.250	-.043	-.191	-.328	-.911	-.685	-.098	-.121	-.246	-.604	-.556
.350	-.104	-.209	-.289	-.710	-.665					
.450	-.145	-.216	-.261	-.480	-.644					
.550	-.157	-.194	-.219	-.322	-.607					
.650	-.116	-.136	-.149	-.201	-.572	-.125	-.080	-.134	-.361	-.475
.750	-.057	-.062	-.070	-.105	-.528	-.069	-.019	-.087	-.323	-.449
.850	-.004	.001	-.005	-.033	-.469	-.025	.023	-.051	-.281	-.425
.900	.020	.031	.028	.036	-.443	-.004	.045	-.026	-.258	-.409
	<i>Right side</i>									
.025	-1.123	-.057	.384	.566	.664	-.999	-.139	.254	.401	.481
.075	-.581	-.136	.172	.360	.505	-.592	-.219	-.026	.130	.260
.150	-.389	-.145	.075	.233	.373	-.284	-.169	-.101	-.020	.093
.250	-.331	-.180	-.017	.117	.239					
.350	-.298	-.198	-.081	.024	.134					
.450	-.260	-.203	-.120	-.040	.047					
.550	-.211	-.180	-.122	-.070	.012					
.650	-.143	-.122	-.090	-.061	-.035	-.105	-.062	-.098	-.125	-.175
.750	-.064	-.043	-.045	-.028	-.053	-.076	-.022	-.060	-.114	-.196
.850	.020	.037	.034	.025	-.052	-.017	.043	.004	-.069	-.190
.900	.058	.064	.057	.045	-.088	.011	.060	.020	-.070	-.213

TABLE VI. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(h)  $\alpha = 9.7^\circ$ ;  $M = 0.85$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.339	.878	.424	-.240	-.021	.464	.739	-.509	-.009	-.855
.025	-.068	-.115	-.103	-.534	-1.479	.324	-.107	-1.180	-1.818	-1.064
.075	.013	-.132	-.308	-.558	-1.070	.149	-.154	-.468	-1.454	-1.083
.150	-.024	-.175	-.387	-.605	-.751	.055	-.161	-.381	-.711	-1.017
.250	-.032	-.121	-.266	-.434	-.508	-.019	-.178	-.328	-.478	-.932
.350	-.062	-.136	-.243	-.335	-.433	-.089	-.215	-.317	-.401	-.810
.450	-.089	-.168	-.239	-.290	-.393	-.148	-.246	-.298	-.385	-.673
.550	-.091	-.156	-.206	-.255	-.356	-.138	-.207	-.239	-.306	-.568
.650	-.074	-.131	-.169	-.216	-.328	-.118	-.162	-.176	-.226	-.485
.750	-.059	-.101	-.135	-.172	-.313	-.074	-.097	-.103	-.139	-.406
.850	-.018	-.036	-.059	-.087	-.226	.000	-.006	-.009	-.040	-.301
.900	.019	.011	-.012	-.039	-.170	.042	.044	.042	.011	-.237
	<i>Right side</i>									
.025	-.052	-.016	-.041	.325	.753	-1.069	-.050	.371	.586	.713
.075	-.332	-.135	.025	.289	.492	-.455	-.121	.166	.368	.526
.150	-.365	-.154	.005	.217	.347	-.372	-.139	.069	.236	.381
.250	-.271	-.108	-.022	.159	.252	-.315	-.159	-.004	.137	.259
.350	-.252	-.138	-.055	.077	.163	-.296	-.185	-.055	.061	.166
.450	-.241	-.159	-.085	.009	.078	-.271	-.209	-.106	-.020	.072
.550	-.191	-.135	-.072	-.014	.051	-.226	-.189	-.120	-.054	.019
.650	-.151	-.107	-.052	-.020	.029	-.162	-.139	-.092	-.050	-.008
.750	-.115	-.084	-.035	-.010	.016	-.076	-.065	-.043	-.023	-.017
.850	-.035	-.009	.018	.024	.019	.015	.023	.031	.031	-.008
.900	.005	.021	.036	.037	-.001	.048	.045	.046	.037	-.041
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.336	.676	.368	-.139	-.656	.347	.605	.212	-.268	-.659
.025	.327	-.154	-1.251	-1.392	-.737	.194	-.266	-1.177	-.811	-.574
.075	.153	-.179	-.677	-1.296	-.739	-.034	-.290	-.719	-.785	-.563
.150	.036	-.199	-.461	-1.087	-.726	-.158	-.227	-.435	-.806	-.564
.250	-.051	-.216	-.354	-.838	-.703	-.117	-.141	-.246	-.705	-.578
.350	-.118	-.239	-.306	-.667	-.683					
.450	-.165	-.252	-.280	-.462	-.663					
.550	-.175	-.217	-.230	-.309	-.630					
.650	-.131	-.151	-.157	-.187	-.599	-.125	-.095	-.139	-.388	-.510
.750	-.067	-.075	-.078	-.097	-.558	-.065	-.030	-.088	-.331	-.485
.850	-.007	-.002	-.006	-.026	-.508	-.011	.021	-.051	-.278	-.455
.900	.023	.033	.032	.007	-.489	.000	.044	-.029	-.249	-.445
	<i>Right side</i>									
.025	-1.258	-.061	.378	.564	.664	-1.164	-.155	.246	.401	.483
.075	-.534	-.144	.166	.358	.503	-.701	-.260	-.043	.116	.261
.150	-.402	-.152	.069	.226	.364	-.310	-.192	-.130	-.052	.082
.250	-.349	-.195	-.029	.107	.232					
.350	-.310	-.227	-.103	.006	.117					
.450	-.268	-.232	-.142	-.063	.025					
.550	-.216	-.200	-.145	-.099	-.043					
.650	-.145	-.136	-.109	-.083	-.070	-.101	-.074	-.100	-.150	-.214
.750	-.065	-.061	-.052	-.052	-.089	-.071	-.033	-.065	-.127	-.232
.850	.030	.033	.036	.017	-.086	-.003	.043	.002	-.077	-.221
.900	.065	.062	.061	.033	-.123	.010	.061	.022	-.074	-.244

TABLE VI. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL  
TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $\alpha = 0^\circ$  - Continued

(i)  $\alpha = 9.7^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.298	.908	.421	-.149	.109	.527	.750	.571	.147	-.478
.025	-.095	-.093	-.019	-.453	-1.313	.324	-.098	-1.041	-1.377	-.977
.075	.022	-.122	-.249	-.490	-.976	.153	-.154	-.523	-1.230	-.992
.150	-.005	-.178	-.411	-.587	-.728	.055	-.162	-.463	-.740	-.955
.250	-.024	-.121	-.289	-.505	-.606	-.018	-.181	-.337	-.645	-.883
.350	-.060	-.142	-.255	-.405	-.506	-.095	-.229	-.355	-.466	-.798
.450	-.095	-.178	-.269	-.322	-.437	-.164	-.272	-.345	-.308	-.682
.550	-.101	-.173	-.221	-.245	-.381	-.163	-.236	-.237	-.305	-.588
.650	-.089	-.146	-.171	-.210	-.339	-.144	-.181	-.181	-.253	-.521
.750	-.071	-.109	-.134	-.179	-.342	-.088	-.104	-.103	-.152	-.451
.850	-.020	-.033	-.062	-.094	-.281	-.004	-.003	-.003	-.040	-.354
.900	.015	.011	-.006	-.039	-.204	.040	.047	.049	.016	-.288
	<i>Right side</i>									
.025	.039	.001	-.054	.320	.751	-.969	-.042	.375	.587	.724
.075	-.271	-.126	.034	.295	.503	-.568	-.121	.173	.368	.538
.150	-.408	-.158	.021	.230	.364	-.424	-.145	.074	.238	.394
.250	-.293	-.116	-.007	.166	.268	-.327	-.162	.002	.139	.273
.350	-.276	-.146	-.050	.080	.171	-.341	-.202	-.062	.057	.176
.450	-.274	-.174	-.079	.007	.082	-.310	-.233	-.119	-.027	.078
.550	-.208	-.153	-.077	-.013	.051	-.234	-.214	-.139	-.063	.016
.650	-.155	-.112	-.057	-.011	.030	-.168	-.160	-.114	-.064	-.013
.750	-.119	-.088	-.045	-.019	.006	-.080	-.069	-.053	-.031	-.025
.850	-.039	-.007	.011	.020	.003	.018	.023	.027	.024	-.025
.900	.006	.025	.033	.030	-.019	.052	.046	.043	.031	-.059
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.396	.688	.438	.006	-.532	.383	.607	.279	-.143	-.604
.025	.321	-.145	-1.179	-1.417	-.711	.178	-.274	-1.137	-.898	-.565
.075	.149	-.178	-.997	-1.311	-.721	-.056	-.331	-1.118	-.837	-.553
.150	.031	-.210	-.421	-1.130	-.717	-.262	-.248	-.511	-.828	-.553
.250	-.064	-.234	-.413	-.795	-.701	-.153	-.136	-.195	-.744	-.567
.350	-.144	-.281	-.378	-.639	-.684					
.450	-.214	-.285	-.249	-.523	-.658					
.550	-.220	-.246	-.233	-.397	-.630					
.650	-.151	-.158	-.157	-.253	-.593	-.137	-.093	-.147	-.412	-.513
.750	-.077	-.076	-.073	-.126	-.561	-.073	-.021	-.093	-.357	-.492
.850	-.009	.006	.005	-.028	-.515	-.021	.034	-.051	-.292	-.467
.900	.023	.039	.039	.011	-.498	-.002	.051	-.029	-.254	-.453
	<i>Right side</i>									
.025	-1.156	-.062	.371	.561	.676	-1.184	-.145	.241	.403	.499
.075	-.938	-.144	.165	.350	.509	-1.061	-.332	-.065	.117	.273
.150	-.358	-.158	.066	.225	.372	-.401	-.212	-.202	-.070	.089
.250	-.409	-.210	-.038	.105	.242					
.350	-.371	-.253	-.123	-.001	.124					
.450	-.262	-.265	-.181	-.089	.024					
.550	-.235	-.228	-.179	-.129	-.047					
.650	-.155	-.150	-.127	-.112	-.082	-.115	-.077	-.113	-.160	-.235
.750	-.072	-.060	-.059	-.063	-.106	-.080	-.026	-.073	-.134	-.245
.850	.028	.041	.033	.012	-.094	-.012	.049	.007	-.077	-.226
.900	.064	.071	.059	.032	-.128	.011	.065	.026	-.070	-.245

TABLE VI. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(j)  $\alpha = 9.7^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.308	.971	.451	-.123		.562	.742	.593	.175	
.025	-.065	-.053	.041	-.403		.330	-.049	-.989	-1.369	
.075	.040	-.077	-.166	-.447		.162	-.108	-.419	-1.227	
.150	.023	-.140	-.329	-.555		.068	-.130	-.372	-.734	
.250	.004	-.102	-.243	-.516		-.009	-.155	-.317	-.658	
.350	-.040	-.132	-.235	-.431		-.084	-.208	-.347	-.545	
.450	-.078	-.178	-.261	-.355		-.153	-.263	-.370	-.308	
.550	-.095	-.188	-.232	-.252		-.172	-.251	-.249	-.299	
.650	-.089	-.172	-.184	-.206		-.151	-.205	-.188	-.259	
.750	-.073	-.145	-.151	-.169		-.094	-.119	-.115	-.162	
.850	-.024	-.065	-.073	-.096		-.002	-.012	-.013	-.045	
.900	.010	-.008	-.015	-.043		.037	-.042	.040	.010	
	<i>Right side</i>									
.025	.114	.044	.004	.302		-.860	-.007	.393	.592	
.075	-.186	-.083	.066	.320		-.465	-.084	.193	.375	
.150	-.332	-.123	.059	.256		-.374	-.111	.096	.246	
.250	-.256	-.095	.033	.179		-.303	-.144	.016	.141	
.350	-.260	-.134	-.010	.092		-.337	-.189	-.052	.055	
.450	-.273	-.176	-.056	.012		-.337	-.231	-.113	-.027	
.550	-.216	-.170	-.069	-.009		-.236	-.226	-.147	-.070	
.650	-.160	-.134	-.060	-.009		-.173	-.176	-.122	-.079	
.750	-.126	-.116	-.048	-.020		-.084	-.087	-.061	-.043	
.850	-.042	-.027	.006	.018		.011	.015	.020	.016	
.900	-.005	.006	.027	.023		.048	.042	.034	.022	
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.413	.679	.438	.036		.411	.610	.295	-.108	
.025	.324	-.104	-1.126	-1.447		.186	-.245	-1.120	-.822	
.075	.157	-.142	-.930	-1.318		-.044	-.301	-1.077	-.816	
.150	.040	-.176	-.402	-1.153		-.249	-.243	-.507	-.826	
.250	-.057	-.214	-.409	-.813		-.152	-.134	-.192	-.784	
.350	-.141	-.280	-.430	-.640						
.450	-.215	-.300	-.258	-.542						
.550	-.219	-.266	-.231	-.431						
.650	-.156	-.171	-.167	-.279		-.137	-.102	-.150	-.431	
.750	-.078	-.078	-.081	-.148		-.066	-.031	-.095	-.370	
.850	-.006	-.002	-.003	-.045		-.017	.026	-.056	-.309	
.900	.024	.035	.034	-.002		.004	.047	-.031	-.277	
	<i>Right side</i>									
.025	-1.080	-.035	.383	.556		-1.118	-.133	.249	.401	
.075	-.789	-.119	.181	.352		-.977	-.314	-.053	.115	
.150	-.361	-.137	.079	.227		-.363	-.210	-.193	-.067	
.250	-.396	-.199	-.028	.102						
.350	-.420	-.252	-.117	-.004						
.450	-.247	-.273	-.186	-.094						
.550	-.233	-.239	-.197	-.139						
.650	-.151	-.176	-.134	-.127		-.112	-.079	-.115	-.176	
.750	-.068	-.045	-.068	-.076		-.074	-.029	-.074	-.144	
.850	.029	.036	.027	.002		-.005	.049	.000	-.080	
.900	.069	.068	.053	.023		.015	.084	.020	-.082	



TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(k)  $\alpha = 15.6^\circ$ ;  $M = 0.60$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	-.332	.815	.199	.103	-.514	.274	.733	.409	-.402	-.999
.025	-.644	-.158	.296	.112	-.532	.209	-.112	-.350	-.726	-1.090
.075	-.198	-.158	.021	-.089	-.462	.080	-.158	-.319	-.530	-.931
.150	-.085	-.178	-.141	-.206	-.437	.037	-.162	-.280	-.416	-.746
.250	-.042	-.139	-.152	-.195	-.371	-.011	-.169	-.265	-.361	-.600
.350	-.031	-.123	-.175	-.206	-.335	-.054	-.196	-.269	-.323	-.501
.450	-.038	-.142	-.193	-.208	-.297	-.081	-.208	-.253	-.296	-.424
.550	-.047	-.153	-.179	-.193	-.251	-.076	-.185	-.217	-.251	-.355
.650	-.045	-.121	-.143	-.157	-.210	-.072	-.158	-.179	-.206	-.297
.750	-.051	-.110	-.125	-.132	-.170	-.063	-.107	-.118	-.150	-.224
.850	-.035	-.066	-.087	-.089	-.108	-.070	-.046	-.062	-.093	-.154
.900	-.022	-.032	-.055	-.055	-.070	-.178	-.190	-.166	-.177	-.160
<i>Right side</i>										
.025	.324	-.103	-.798	-.298	-.102	-.569	-.110	.112	.197	.200
.075	-.072	-.185	-.229	-.089	.012	-.409	-.160	.024	.121	.214
.150	-.198	-.167	-.075	-.010	.087	-.332	-.162	.012	.089	.216
.250	-.194	-.121	-.015	.035	.132	-.293	-.167	.001	.080	.227
.350	-.203	-.130	-.001	.053	.148	-.271	-.178	-.024	.064	.205
.450	-.212	-.144	-.008	.044	.134	-.259	-.194	-.046	.042	.182
.550	-.189	-.139	-.017	.033	.100	-.219	-.180	-.053	.028	.146
.650	-.149	-.112	-.021	.017	.066	-.173	-.144	-.044	.024	.114
.750	-.119	-.094	-.017	.008	.041	-.112	-.091	-.024	.024	.087
.850	-.069	-.037	-.003	.015	.019	-.047	-.023	.008	.035	.069
.900	-.047	-.014	.003	.010	.012	-.020	-.012	.003	.021	.046
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	-.020	.630	.105	-.636	-.984	.098	.543	-.145	-.620	-.816
.025	.370	-.130	-.963	-1.328	-1.022	.200	-.208	-.949	-.810	-.650
.075	.189	-.160	-.544	-1.199	-1.002	.010	-.190	-.470	-.778	-.650
.150	.078	-.171	-.402	-.893	-.968	-.094	-.171	-.303	-.706	-.637
.250	.001	-.178	-.321	-.517	-.879	-.072	-.114	-.229	-.611	-.600
.350	-.054	-.194	-.287	-.355	-.741					
.450	-.087	-.201	-.262	-.283	-.598					
.550	-.099	-.190	-.226	-.240	-.473					
.650	-.085	-.149	-.172	-.177	-.369	-.110	-.085	-.147	-.343	-.451
.750	-.051	-.094	-.107	-.123	-.287	-.074	-.053	-.123	-.289	-.410
.850	-.015	-.037	-.051	-.075	-.215	-.054	-.014	-.096	-.235	-.369
.900	-.006	-.016	-.024	-.046	-.183	-.035	.004	-.069	-.202	-.342
<i>Right side</i>										
.025	-1.038	-.112	.382	.553	.558	-.811	-.167	.220	.334	.391
.075	-.599	-.162	.166	.348	.431	-.531	-.201	-.006	.100	.202
.150	-.395	-.160	.078	.224	.343	-.339	-.180	-.098	-.042	.055
.250	-.332	-.176	.010	.130	.254					
.350	-.305	-.194	-.042	.066	.191					
.450	-.268	-.201	-.071	.033	.141					
.550	-.228	-.176	-.078	.008	.105					
.650	-.169	-.137	-.055	.008	.084	-.081	-.089	-.084	-.084	-.086
.750	-.106	-.085	-.030	.017	.057	-.121	-.057	-.084	-.093	-.120
.850	-.024	-.005	.021	.044	.039	-.069	.004	-.024	-.053	-.111
.900	.001	.016	.033	.037	.021	-.047	.018	-.010	-.046	-.133

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Continued

(i)  $\alpha = 15.8^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	-.025	.857	.398	.216	-.494	.399	.774	.420	-.203	-.656
.025	-.991	.095	.351	.081	-.460	.004	-.054	-.357	-.600	-.795
.075	-.639	-.105	.005	-.115	-.429	-.078	-.155	-.355	-.502	-.720
.150	-.299	-.209	-.210	-.254	-.429	-.076	-.174	-.323	-.441	-.630
.250	-.144	-.152	-.193	-.219	-.372	-.066	-.187	-.304	-.405	-.561
.350	-.080	-.133	-.208	-.227	-.329	-.076	-.212	-.292	-.372	-.493
.450	-.046	-.147	-.219	-.224	-.287	-.083	-.231	-.278	-.330	-.433
.550	-.035	-.133	-.202	-.195	-.241	-.064	-.208	-.236	-.280	-.365
.650	-.028	-.115	-.157	-.163	-.202	-.055	-.174	-.193	-.233	-.307
.750	-.029	-.114	-.146	-.142	-.165	-.037	-.114	-.133	-.165	-.237
.850	-.022	-.068	-.099	-.089	-.106	-.006	-.041	-.063	-.100	-.170
.900	-.005	-.033	-.064	-.059	-.074	-.172	-.179	-.145	-.147	-.137
<i>Right side</i>										
.025	.420	-.772	-.961	-.517	-.293	-.465	-.293	-.036	.025	.044
.075	-.011	-.422	-.664	-.269	-.176	-.392	-.278	-.105	-.020	.085
.150	-.217	-.257	-.325	-.172	-.064	-.349	-.225	-.092	-.020	.141
.250	-.214	-.152	-.152	-.094	.035	-.310	-.176	-.064	.004	.185
.350	-.226	-.102	-.080	-.038	.096	-.293	-.165	-.057	.013	.199
.450	-.220	-.091	-.046	-.005	.125	-.272	-.174	-.063	.016	.191
.550	-.188	-.085	-.031	.012	.110	-.237	-.155	-.055	.018	.164
.650	-.148	-.068	-.020	.013	.075	-.185	-.117	-.042	.025	.135
.750	-.134	-.062	-.019	.016	.047	-.118	-.068	-.014	.032	.099
.850	-.073	-.024	-.002	.022	.029	-.046	-.003	.016	.044	.075
.900	-.055	-.007	.004	.021	.013	-.023	.008	.010	.029	.041
$z/b_v = 0.66$										
<i>Left side</i>										
.000	.320	.667	.430	.001	-.538	.241	.572	.128	-.346	-.745
.025	.312	-.109	-1.076	-1.314	-.901	.177	-.257	-1.064	-.833	-.628
.075	.143	-.159	-.572	-1.159	-.890	-.011	-.234	-.619	-.784	-.605
.150	.042	-.188	-.436	-.878	-.856	-.115	-.191	-.308	-.710	-.588
.250	-.012	-.202	-.334	-.544	-.792	-.064	-.114	-.223	-.610	-.563
.350	-.060	-.217	-.299	-.385	-.697					
.450	-.093	-.228	-.275	-.296	-.580					
.550	-.102	-.214	-.237	-.239	-.475					
.650	-.076	-.159	-.172	-.171	-.385	-.095	-.070	-.131	-.345	-.415
.750	-.034	-.085	-.099	-.104	-.305	-.064	-.026	-.095	-.281	-.382
.850	.000	-.024	-.028	-.050	-.240	-.031	.012	-.061	-.221	-.349
.900	.013	.002	.002	-.020	-.209	-.012	.034	-.036	-.181	-.330
<i>Right side</i>										
.025	-1.211	-.144	.325	.533	.621	-1.005	-.173	.201	.359	.456
.075	-.590	-.184	.124	.314	.462	-.647	-.226	-.040	.109	.256
.150	-.413	-.153	.051	.202	.367	-.331	-.178	-.125	-.050	.082
.250	-.349	-.168	-.011	.122	.291					
.350	-.310	-.184	-.061	.054	.222					
.450	-.272	-.184	-.083	.022	.164					
.550	-.226	-.162	-.078	.006	.127					
.650	-.162	-.115	-.057	.015	.097	-.115	-.064	-.067	-.071	-.090
.750	-.089	-.057	-.028	.024	.072	-.101	-.030	-.061	-.076	-.110
.850	-.003	.023	.036	.059	.055	-.034	.037	.002	-.026	-.101
.900	.027	.047	.048	.054	.024	-.015	.049	.017	-.024	-.124

TABLE VI.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE WING, FUSELAGE, VERTICAL  
TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE LOW POSITION;  $i_t = 0^\circ$  - Concluded

(m)  $\alpha = 15.9^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	-.357	.905	.565	.243	-.610	.441	.769	.463	-.211	-.589
.025	-1.038	-.009	.281	-.052	-.583	.244	-.183	-.520	-.672	-.764
.075	-.511	-.227	-.041	-.221	-.528	.030	-.267	-.476	-.576	-.699
.150	-.255	-.340	-.300	-.362	-.502	-.062	-.258	-.415	-.524	-.627
.250	-.200	-.237	-.260	-.300	-.438	-.114	-.243	-.362	-.485	-.568
.350	-.183	-.192	-.256	-.284	-.378	-.169	-.267	-.341	-.436	-.509
.450	-.183	-.196	-.252	-.269	-.315	-.204	-.286	-.315	-.381	-.442
.550	-.169	-.183	-.225	-.229	-.258	-.183	-.243	-.271	-.320	-.380
.650	-.132	-.136	-.169	-.187	-.216	-.159	-.200	-.222	-.260	-.323
.750	-.117	-.126	-.162	-.167	-.170	-.115	-.141	-.158	-.188	-.249
.850	-.080	-.079	-.114	-.111	-.112	-.053	-.059	-.085	-.116	-.190
.900	-.041	-.040	-.077	-.073	-.070	-.191	-.195	-.154	-.153	-.137
	<i>Right side</i>									
.025	.234	-.687	-1.129	-.582	-.419	-1.084	-.317	-.033	-.039	-.021
.075	-.204	-.887	-.757	-.356	-.293	-.798	-.360	-.149	-.099	-.011
.150	-.504	-.364	-.371	-.257	-.166	-.680	-.299	-.159	-.099	.062
.250	-.491	-.262	-.234	-.184	-.053	-.377	-.251	-.136	-.066	.119
.350	-.286	-.195	-.169	-.116	.026	-.307	-.234	-.125	-.053	.143
.450	-.252	-.161	-.117	-.068	.072	-.294	-.237	-.125	-.042	.146
.550	-.215	-.127	-.081	-.036	.085	-.256	-.203	-.104	-.025	.136
.650	-.160	-.096	-.057	-.019	.071	-.197	-.152	-.074	-.007	.119
.750	-.134	-.079	-.040	-.005	.052	-.121	-.096	-.044	.008	.093
.850	-.070	-.037	-.013	.008	.034	-.046	-.021	-.002	.025	.069
.900	-.048	-.024	-.007	.003	.022	-.008	-.011	-.010	.012	.042
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.297	.657	.452	.115	-.341	.255	.567	.179	-.294	-.687
.025	.289	-.172	-1.226	-1.427	-.858	.080	-.302	-1.234	-.983	-.580
.075	.109	-.217	-.685	-1.235	-.844	-.075	-.315	-.732	-.903	-.585
.150	-.013	-.240	-.497	-.873	-.820	-.228	-.237	-.377	-.794	-.576
.250	-.089	-.244	-.384	-.623	-.773	-.117	-.144	-.246	-.679	-.548
.350	-.155	-.261	-.324	-.473	-.693					
.450	-.198	-.272	-.298	-.361	-.590					
.550	-.204	-.253	-.263	-.270	-.489					
.650	-.160	-.185	-.189	-.191	-.395	-.131	-.090	-.148	-.356	-.418
.750	-.094	-.106	-.111	-.117	-.318	-.097	-.045	-.109	-.294	-.387
.850	-.042	-.040	-.037	-.056	-.253	-.051	.006	-.071	-.232	-.353
.900	-.015	-.007	-.007	-.026	-.220	-.022	.024	-.046	-.192	-.327
	<i>Right side</i>									
.025	-1.253	-.186	.284	.506	.646	-1.256	-.234	.169	.349	.478
.075	-1.032	-.243	.077	.277	.469	-.866	-.308	-.090	.090	.270
.150	-.632	-.215	.001	.168	.367	-.506	-.231	-.188	-.094	.091
.250	-.301	-.229	-.061	.081	.283					
.350	-.289	-.243	-.117	.019	.213					
.450	-.276	-.234	-.134	-.019	.169					
.550	-.242	-.202	-.124	-.028	.129					
.650	-.176	-.147	-.094	-.014	.102	-.096	-.081	-.104	-.096	-.060
.750	-.104	-.082	-.051	.008	.073	-.110	-.048	-.084	-.092	-.110
.850	-.014	.011	.020	.044	.058	-.045	.028	-.011	-.038	-.099
.900	.024	.035	.034	.046	.024	-.017	.039	.001	-.036	-.123

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUELAGES AND VERTICAL TAIL

(a)  $\alpha = 0^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.642	.645	.338	-.425	-1.270	.270	.728	.329	-.416	-1.192
.025	.263	-.113	-.543	-1.130	-2.267	.330	-.107	-.623	-1.079	-1.133
.075	.123	-.120	-.369	-.669	-1.505	.130	-.143	-.431	-.939	-1.105
.150	.036	-.146	-.302	-.476	-.667	.050	-.146	-.318	-.697	-1.057
.250	.013	-.107	-.204	-.303	-.416	-.017	-.152	-.263	-.425	-1.005
.350	-.021	-.111	-.167	-.236	-.336	-.072	-.171	-.245	-.328	-.843
.450	-.067	-.129	-.156	-.199	-.281	-.104	-.175	-.222	-.266	-.639
.550	-.069	-.107	-.112	-.146	-.298	-.095	-.143	-.161	-.195	-.477
.650	-.040	-.065	-.062	-.093	-.210	-.067	-.100	-.108	-.137	-.352
.750	-.010	-.021	-.016	-.047	-.123	-.028	-.040	-.039	-.065	-.220
.850	.032	.029	.036	-.003	-.037	.036	.025	.029	.011	-.117
.900	.055	.059	.066	.031	.009	.064	.061	.068	.050	-.057
	<i>Right side</i>									
.025	-.452	-.035	.306	.580	.815	-.620	-.063	.364	.591	.735
.075	-.374	-.118	.123	.333	.536	-.418	-.125	.148	.349	.516
.150	-.294	-.127	.043	.204	.374	-.310	-.134	.055	.209	.361
.250	-.205	-.104	.013	.130	.265	-.250	-.146	-.014	.105	.230
.350	-.179	-.120	-.035	.057	.164	-.228	-.159	-.062	.036	.146
.450	-.159	-.132	-.074	.001	.091	-.200	-.162	-.094	-.012	.071
.550	-.120	-.109	-.064	-.017	.057	-.160	-.141	-.090	-.036	.027
.650	-.065	-.061	-.039	.001	.050	-.100	-.100	-.064	-.024	.023
.750	-.012	-.017	-.009	.020	.052	-.030	-.031	-.019	.006	.027
.850	.052	.043	.045	.059	.080	.040	.038	.055	.054	.055
.900	.077	.064	.061	.066	.073	.060	.059	.064	.059	.039
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.240	.716	.279	-.388	-.660	.380	.668	.240	-.284	-.438
.025	.337	-.141	-.808	-1.054	-.726	.180	-.205	-.652	-.860	-.518
.075	.151	-.159	-.481	-.953	-.710	.010	-.164	-.371	-.803	-.518
.150	.036	-.169	-.360	-.780	-.696	-.040	-.148	-.273	-.701	-.553
.250	-.030	-.171	-.279	-.529	-.664	-.030	-.113	-.213	-.559	-.546
.350	-.081	-.182	-.247	-.360	-.633					
.450	-.111	-.185	-.222	-.275	-.601					
.550	-.111	-.162	-.181	-.211	-.555					
.650	-.083	-.118	-.117	-.151	-.500	-.060	-.070	-.096	-.365	-.391
.750	-.044	-.049	-.046	-.079	-.432	-.050	-.038	-.071	-.342	-.363
.850	.018	.006	.009	-.019	-.365	-.020	-.008	-.048	-.294	-.336
.900	.036	.034	.036	.008	-.327	-.000	.018	-.028	-.257	-.322
	<i>Right side</i>									
.025	-.705	-.074	.380	.584	.683	-.550	-.141	.219	.368	.441
.075	-.466	-.141	.148	.345	.486	-.380	-.164	.009	.110	.208
.150	-.329	-.146	.055	.206	.329	-.260	-.155	-.046	.006	.082
.250	-.278	-.166	-.025	.093	.199					
.350	-.255	-.180	-.083	.013	.098					
.450	-.218	-.169	-.108	-.033	.032					
.550	-.170	-.150	-.110	-.052	-.005					
.650	-.108	-.107	-.074	-.040	-.016	-.070	-.065	-.058	-.068	-.080
.750	-.044	-.047	-.030	-.012	-.028	-.060	-.047	-.062	-.072	-.110
.850	.029	.025	.032	.036	-.025	-.010	.011	.000	-.024	-.107
.900	.059	.057	.064	.050	-.055	-.000	.025	.000	-.017	-.137

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(b)  $\alpha = 0^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	<i>C<sub>p</sub> for -</i>					<i>C<sub>p</sub> for -</i>				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.745	.655	.520	.212	-.247	.479	.761	.548	.061	-.547
.025	.268	-.099	-.546	-1.127	-1.405	.319	-.099	-.935	-1.656	-.950
.075	.134	-.126	-.378	-.875	-1.191	.140	-.159	-.455	-1.469	-.962
.150	.035	-.171	-.333	-.567	-.988	.046	-.171	-.346	-.486	-.930
.250	.011	-.134	-.223	-.314	-.708	-.026	-.183	-.284	-.355	-.842
.350	-.033	-.143	-.187	-.241	-.486	-.085	-.205	-.269	-.323	-.738
.450	-.079	-.165	-.175	-.206	-.386	-.133	-.223	-.240	-.273	-.623
.550	-.085	-.143	-.127	-.144	-.293	-.118	-.182	-.175	-.188	-.513
.650	-.053	-.093	-.065	-.094	-.190	-.083	-.130	-.107	-.111	-.425
.750	-.018	-.040	-.007	-.036	-.132	-.030	-.053	-.030	-.029	-.326
.850	.032	.023	.053	.022	-.059	.041	.029	.050	.049	-.225
.900	.064	.055	.083	.060	-.009	.079	.066	.090	.085	-.161
	<i>Right side</i>									
.025	-.412	-.026	.314	.581	.815	-.921	-.099	.361	.587	.743
.075	-.396	-.131	.131	.344	.553	-.427	-.156	.152	.354	.532
.150	-.311	-.146	.048	.212	.382	-.336	-.160	.053	.213	.377
.250	-.220	-.134	.013	.142	.277	-.272	-.176	-.017	.113	.254
.350	-.190	-.146	-.040	.060	.170	-.249	-.197	-.074	.032	.152
.450	-.173	-.165	-.088	-.007	.081	-.223	-.205	-.117	-.029	.057
.550	-.123	-.134	-.088	-.026	.038	-.171	-.174	-.118	-.056	.008
.650	-.057	-.082	-.052	-.012	.028	-.094	-.120	-.077	-.036	-.004
.750	.002	-.031	-.017	.009	.025	-.013	-.036	-.016	.005	.002
.850	.064	.040	.047	.060	.052	.064	.044	.061	.063	.022
.900	.090	.061	.067	.069	.055	.096	.061	.076	.072	.003
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.444	.743	.491	.020	-.390	.477	.689	.384	-.095	-.287
.025	.332	-.136	-1.098	-1.626	-.714	.142	-.236	-1.052	-1.237	-.494
.075	.149	-.177	-.555	-1.320	-.711	-.009	-.210	-.486	-1.038	-.469
.150	.025	-.199	-.400	-.978	-.714	-.077	-.186	-.283	-.805	-.504
.250	-.048	-.208	-.306	-.655	-.671	-.047	-.139	-.234	-.592	-.506
.350	-.106	-.223	-.263	-.395	-.633					
.450	-.141	-.226	-.234	-.247	-.582					
.550	-.130	-.197	-.185	-.170	-.521					
.650	-.091	-.131	-.113	-.095	-.472	-.062	-.077	-.089	-.320	-.387
.750	-.042	-.057	-.034	-.032	-.415	-.045	-.040	-.055	-.293	-.357
.850	.034	.012	.033	.025	-.355	-.006	.006	-.028	-.249	-.333
.900	.057	.041	.064	.052	-.323	.020	.029	-.010	-.200	-.317
	<i>Right side</i>									
.025	-1.179	-.105	.377	.588	.710	-1.073	-.200	.201	.368	.479
.075	-.501	-.180	.152	.350	.508	-.421	-.216	-.023	.090	.225
.150	-.351	-.176	.053	.212	.356	-.266	-.188	-.068	-.023	.081
.250	-.299	-.203	-.031	.091	.213					
.350	-.264	-.226	-.104	-.004	.102					
.450	-.223	-.216	-.132	-.054	.025					
.550	-.168	-.186	-.129	-.076	-.026					
.650	-.097	-.125	-.089	-.050	-.042	-.061	-.063	-.055	-.056	-.083
.750	-.023	-.050	-.030	-.015	-.050	-.042	-.043	-.045	-.056	-.123
.850	.061	.040	.053	.050	-.044	.020	.027	.021	-.001	-.109
.900	.095	.069	.079	.069	-.070	.026	.043	.033	-.009	-.140

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(c)  $\alpha = 0^\circ$ ;  $M = 0.85$

$x$ $C_V$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.772	.676	.578	.331	-.059	.529	.776	.612	.213	-.343
.025	.270	-.070	-.525	-1.034	-1.166	.322	-.078	-1.062	-1.446	-.861
.075	.136	-.103	-.381	-.824	-1.025	.139	-.142	-.434	-1.290	-.868
.150	.034	-.152	-.353	-.676	-.895	.046	-.159	-.357	-1.146	-.861
.250	.004	-.122	-.236	-.374	-.729	-.031	-.176	-.304	-.368	-.773
.350	-.043	-.136	-.199	-.266	-.575	-.100	-.206	-.281	-.288	-.711
.450	-.096	-.162	-.187	-.203	-.467	-.158	-.226	-.253	-.223	-.617
.550	-.101	-.136	-.132	-.135	-.357	-.141	-.183	-.177	-.151	-.525
.650	-.067	-.085	-.069	-.084	-.251	-.097	-.125	-.107	-.087	-.449
.750	-.030	-.029	-.002	-.023	-.179	-.047	-.042	-.029	-.013	-.359
.850	.030	.041	.063	.037	-.094	.042	.045	.059	.068	-.268
.900	.062	.078	.092	.072	-.036	.083	.091	.099	.105	-.202
	<i>Right side</i>									
.025	-.386	-.013	.316	.584	.814	-.950	-.089	.357	.582	.743
.075	-.400	-.119	.130	.345	.555	-.434	-.147	.149	.351	.530
.150	-.328	-.140	.049	.216	.388	-.351	-.159	.051	.216	.376
.250	-.237	-.127	.012	.143	.279	-.291	-.176	-.025	.111	.248
.350	-.206	-.145	-.046	.055	.172	-.266	-.199	-.088	.023	.145
.450	-.186	-.163	-.100	-.020	.070	-.234	-.206	-.135	-.050	.043
.550	-.126	-.136	-.106	-.043	.023	-.174	-.176	-.137	-.074	-.013
.650	-.060	-.076	-.063	-.024	.009	-.094	-.115	-.095	-.054	-.026
.750	.002	-.025	-.025	.000	.004	-.011	-.023	-.023	-.003	-.018
.850	.073	.053	.046	.057	.033	.074	.063	.062	.064	-.001
.900	.096	.079	.068	.069	.039	.099	.082	.076	.079	-.020
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.492	.761	.555	.169	-.252	.500	.699	.430	.037	-.226
.025	.334	-.115	-1.153	-1.541	-.702	.137	-.234	-1.146	-1.374	-.497
.075	.150	-.163	-.629	-1.382	-.698	-.037	-.216	-.667	-1.124	-.479
.150	.026	-.192	-.417	-1.234	-.712	-.103	-.184	-.249	-.990	-.517
.250	-.061	-.204	-.313	-.634	-.650	-.063	-.130	-.241	-.611	-.513
.350	-.127	-.226	-.273	-.226	-.624					
.450	-.164	-.230	-.247	-.189	-.583					
.550	-.157	-.193	-.189	-.152	-.530					
.650	-.103	-.122	-.109	-.091	-.479	-.071	-.065	-.092	-.323	-.404
.750	-.054	-.039	-.033	-.027	-.421	-.044	-.022	-.059	-.294	-.378
.850	.032	.034	.038	.034	-.360	-.003	.028	-.028	-.253	-.346
.900	.034	.061	.069	.065	-.330	.020	.053	-.012	-.205	-.329
	<i>Right side</i>									
.025	-1.144	-.100	.370	.578	.711	-1.140	-.207	.183	.356	.482
.075	-.460	-.172	.147	.346	.509	-.616	-.230	-.059	.068	.223
.150	-.373	-.170	.045	.212	.353	-.217	-.187	-.100	-.070	.074
.250	-.311	-.204	-.051	.087	.213					
.350	-.276	-.230	-.126	-.023	.095					
.450	-.234	-.219	-.157	-.075	.008					
.550	-.178	-.183	-.147	-.095	-.047					
.650	-.101	-.109	-.098	-.065	-.062	-.074	-.055	-.060	-.058	-.100
.750	-.023	-.029	-.035	-.021	-.070	-.050	-.025	-.049	-.056	-.139
.850	.070	.062	.056	.060	-.054	.020	.048	.025	.003	-.118
.900	.097	.093	.083	.078	-.081	.023	.062	.034	.006	-.149

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(d)  $\alpha = 0^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.801	.685	.616	.418	.076	.578	.792	.664	.316	-.200
.025	.278	-.064	-.495	-.981	-1.063	.319	-.086	-.931	-1.300	-.781
.075	.147	-.096	-.399	-.785	-.935	.140	-.153	-.677	-1.182	-.798
.150	.035	-.160	-.395	-.690	-.825	.043	-.172	-.427	-1.085	-.776
.250	.006	-.129	-.274	-.548	-.691	-.032	-.185	-.338	-.668	-.722
.350	-.046	-.147	-.232	-.362	-.585	-.112	-.223	-.356	-.492	-.679
.450	-.114	-.180	-.227	-.284	-.515	-.181	-.259	-.290	-.281	-.622
.550	-.123	-.153	-.154	-.161	-.432	-.168	-.204	-.189	-.122	-.555
.650	-.087	-.088	-.076	-.082	-.339	-.118	-.131	-.112	-.066	-.491
.750	-.038	-.026	-.010	-.020	-.250	-.045	-.036	-.030	-.004	-.411
.850	.029	.049	.054	.035	-.159	.041	.056	.056	.071	-.335
.900	.064	.085	.086	.069	-.100	.090	.105	.105	.109	-.281
<i>Right side</i>										
.025	-.353	.013	.311	.584	.821	-.899	-.049	.334	.575	.745
.075	-.424	-.100	.122	.343	.561	-.632	-.137	.129	.340	.528
.150	-.362	-.137	.038	.209	.391	-.392	-.156	.032	.201	.375
.250	-.262	-.126	-.008	.136	.281	-.314	-.180	-.046	.092	.248
.350	-.231	-.150	-.068	.042	.170	-.311	-.215	-.117	.000	.139
.450	-.208	-.177	-.128	-.040	.060	-.246	-.232	-.181	-.082	.030
.550	-.135	-.147	-.140	-.075	.005	-.176	-.193	-.183	-.115	-.034
.650	-.060	-.082	-.096	-.052	-.010	-.091	-.117	-.131	-.090	-.057
.750	.008	-.021	-.049	-.032	-.028	.000	-.016	-.042	-.032	-.053
.850	.075	.060	.034	.041	.001	.085	.073	.051	.048	-.035
.900	.102	.089	.055	.056	-.007	.112	.094	.069	.060	-.067
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.542	.777	.614	.271	-.129	.533	.711	.478	.144	-.160
.025	.329	-.123	-1.045	-1.385	-.677	.120	-.307	-.993	-1.341	-.461
.075	.147	-.174	-.883	-1.255	-.681	-.072	-.276	-.965	-1.285	-.464
.150	.020	-.208	-.481	-1.165	-.696	-.179	-.208	-.489	-1.122	-.499
.250	-.075	-.232	-.414	-1.109	-.629	-.075	-.142	-.179	-.979	-.512
.350	-.153	-.259	-.287	-.676	-.610					
.450	-.199	-.258	-.235	-.284	-.575					
.550	-.179	-.209	-.201	-.093	-.534					
.650	-.123	-.118	-.116	-.032	-.496	-.069	-.057	-.100	-.241	-.425
.750	-.044	-.028	-.031	.004	-.456	-.034	-.014	-.059	-.260	-.400
.850	.036	.049	.043	.048	-.404	.010	.039	-.034	-.241	-.373
.900	.063	.078	.081	.069	-.380	.033	.065	-.011	-.201	-.362
<i>Right side</i>										
.025	-1.033	-.065	.348	.563	.710	-1.068	-.199	.152	.337	.485
.075	-.833	-.158	.124	.330	.502	-.915	-.270	-.116	.038	.221
.150	-.366	-.165	.024	.199	.352	-.278	-.208	-.202	-.124	.063
.250	-.390	-.221	-.081	.067	.209					
.350	-.239	-.254	-.179	-.061	.084					
.450	-.234	-.247	-.219	-.138	-.019					
.550	-.180	-.196	-.197	-.157	-.086					
.650	-.092	-.111	-.130	-.111	-.100	-.063	-.048	-.070	-.084	-.132
.750	-.010	-.020	-.049	-.050	-.102	-.040	-.013	-.057	-.073	-.171
.850	.082	.076	.050	.041	-.077	.031	.065	.021	-.006	-.141
.900	.114	.107	.082	.063	-.114	.037	.077	.034	-.011	-.180

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(e)  $\alpha = 0^\circ$ ;  $M = 0.92$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.813	.684	.631	.448	.112	.600	.796	.682	.357	-.160
.025	.272	-.059	-.471	-.928	-1.031	.311	-.081	-.916	-1.233	-.733
.075	.140	-.101	-.387	-.738	-.896	.131	-.159	-.710	-1.125	-.746
.150	.028	-.167	-.391	-.664	-.785	.034	-.177	-.432	-1.034	-.735
.250	-.005	-.139	-.289	-.566	-.660	-.045	-.197	-.340	-.727	-.700
.350	-.056	-.160	-.232	-.382	-.568	-.126	-.246	-.358	-.548	-.659
.450	-.127	-.204	-.231	-.316	-.514	-.207	-.297	-.334	-.415	-.601
.550	-.142	-.180	-.150	-.175	-.440	-.196	-.230	-.176	-.110	-.539
.650	-.102	-.104	-.066	-.073	-.358	-.135	-.133	-.098	-.031	-.478
.750	-.049	-.031	.002	-.005	-.268	-.056	-.037	-.017	.023	-.407
.850	.023	.049	.064	.050	-.173	.036	.057	.068	.090	-.338
.900	.058	.087	.096	.085	-.111	.085	.104	.110	.123	-.277
	<i>Right side</i>									
.025	-.335	.005	.323	.591	.832	-.841	-.068	.346	.583	.754
.075	-.420	-.113	.135	.352	.568	-.656	-.151	.141	.348	.539
.150	-.380	-.142	.049	.221	.405	-.432	-.167	.039	.215	.385
.250	-.289	-.138	.002	.145	.290	-.326	-.193	-.046	.104	.256
.350	-.246	-.168	-.061	.052	.182	-.338	-.234	-.115	.009	.150
.450	-.225	-.198	-.123	-.032	.074	-.293	-.273	-.184	-.081	.037
.550	-.144	-.176	-.144	-.072	.012	-.171	-.217	-.191	-.118	-.029
.650	-.060	-.097	-.095	-.054	-.008	-.089	-.122	-.131	-.094	-.056
.750	.003	-.026	-.041	-.028	-.028	-.003	-.014	-.033	-.024	-.053
.850	.074	.055	.041	.050	-.002	.085	.074	.061	.059	-.039
.900	.097	.087	.061	.068	-.012	.110	.096	.076	.075	-.064
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.559	.778	.631	.314	-.102	.547	.712	.498	.191	-.140
.025	.317	-.120	-1.022	-1.312	-.677	.090	-.297	-.997	-1.282	-.488
.075	.140	-.177	-.877	-1.186	-.677	-.085	-.306	-.955	-1.224	-.454
.150	.008	-.217	-.674	-1.112	-.702	-.233	-.258	-.740	-1.076	-.486
.250	-.090	-.252	-.399	-1.072	-.616	-.122	-.151	-.199	-.949	-.502
.350	-.184	-.318	-.404	-.962	-.594					
.450	-.238	-.292	-.186	-.371	-.565					
.550	-.203	-.231	-.176	-.064	-.525					
.650	-.142	-.117	-.104	.002	-.494	-.074	-.059	-.094	-.289	-.417
.750	-.036	-.026	-.020	.038	-.456	-.041	-.009	-.057	-.269	-.397
.850	.034	.050	.055	.077	-.409	.005	.042	-.030	-.237	-.371
.900	.061	.082	.089	.098	-.379	.027	.071	-.012	-.203	-.357
	<i>Right side</i>									
.025	-.971	-.083	.354	.571	.716	-1.013	-.217	.159	.348	.490
.075	-.820	-.175	.134	.341	.510	-.914	-.337	-.112	.053	.230
.150	-.543	-.180	.031	.207	.357	-.616	-.252	-.223	-.114	.070
.250	-.393	-.247	-.080	.069	.217					
.350	-.376	-.321	-.186	-.057	.085					
.450	-.193	-.279	-.232	-.142	-.015					
.550	-.173	-.218	-.201	-.167	-.083					
.650	-.098	-.109	-.124	-.112	-.102	-.069	-.048	-.064	-.068	-.122
.750	-.016	-.022	-.042	-.040	-.106	-.048	-.016	-.051	-.057	-.168
.850	.081	.078	.061	.056	-.077	.027	.065	.033	.010	-.136
.900	.111	.109	.089	.085	-.105	.028	.083	.043	.009	-.177



TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(f)  $\alpha = 9^\circ$ ;  $M = 0.60$

$\frac{x}{C_V}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_V = 0.11$					$z/b_V = 0.38$				
	<i>Left side</i>									
.000	.419	.828	.555	.172	-1.068	.075	.650	.007	-.980	-1.806
.025	-.142	-.122	-.023	-.426	-.993	.322	-.075	-.692	-1.297	-1.638
.075	.001	-.118	-.155	-.389	-.694	.159	-.093	-.386	-1.364	-2.051
.150	.014	-.136	-.196	-.352	-.514	.078	-.091	-.267	-.385	-1.631
.250	.012	-.104	-.139	-.251	-.346	.017	-.095	-.219	-.294	-.397
.350	-.013	-.100	-.139	-.207	-.285	-.029	-.120	-.208	-.271	-.355
.450	-.035	-.111	-.139	-.186	-.248	-.065	-.127	-.180	-.234	-.322
.550	-.047	-.104	-.119	-.151	-.210	-.054	-.093	-.139	-.179	-.259
.650	-.022	-.063	-.073	-.098	-.154	-.033	-.063	-.094	-.131	-.213
.750	-.008	-.036	-.037	-.040	-.098	-.002	-.018	-.039	-.075	-.161
.850	.023	.009	.014	.024	-.007	.044	.053	.027	-.001	-.079
.900	.055	.043	.041	.059	.040	.064	.089	.059	.033	-.030
	<i>Right side</i>									
.025	.048	-.063	-.062	.193	.678	-.563	.032	.370	.530	.624
.075	-.158	-.116	.009	.172	.516	-.354	-.034	.185	.348	.481
.150	-.189	-.104	.030	.144	.390	-.262	-.052	.096	.223	.360
.250	-.149	-.084	.027	.156	.308	-.216	-.068	.036	.133	.241
.350	-.146	-.093	.000	.121	.217	-.192	-.084	-.005	.075	.175
.450	-.146	-.109	-.028	.080	.152	-.173	-.093	-.041	.024	.101
.550	-.121	-.097	-.032	.052	.115	-.140	-.082	-.043	.003	.068
.650	-.074	-.063	-.012	.047	.101	-.087	-.052	-.025	.006	.054
.750	-.042	-.027	.007	.052	.096	-.029	.009	.011	.029	.061
.850	.028	.023	.048	.091	.115	.035	.068	.062	.070	.082
.900	.046	.043	.064	.093	.119	.062	.091	.071	.073	.068
	$z/b_V = 0.66$					$z/b_V = 0.93$				
	<i>Left side</i>									
.000	.028	.582	-.059	-.736	-1.248	.182	.523	-.066	-.410	-.790
.025	.297	-.122	-.886	-1.057	-1.000	.143	-.204	-.731	-.629	-.603
.075	.141	-.129	-.441	-1.013	-.998	-.017	-.166	-.384	-.602	-.596
.150	.044	-.129	-.320	-.969	-1.002	-.094	-.132	-.235	-.551	-.577
.250	-.020	-.122	-.242	-.713	-1.026	-.065	-.070	-.149	-.458	-.547
.350	-.065	-.129	-.217	-.338	-.923					
.450	-.087	-.129	-.190	-.191	-.741					
.550	-.092	-.118	-.153	-.165	-.561					
.650	-.067	-.070	-.098	-.112	-.374	-.072	-.020	-.073	-.276	-.463
.750	-.026	-.009	-.034	-.050	-.208	-.035	.009	-.041	-.237	-.453
.850	.010	.028	.014	-.003	-.110	-.006	.043	-.007	-.186	-.425
.900	.028	.059	.041	.024	-.065	.010	.064	.009	-.158	-.399
	<i>Right side</i>									
.025	-.671	.000	.354	.502	.575	-.542	-.084	.187	.278	.343
.075	-.404	-.066	.162	.322	.444	-.339	-.134	-.007	.066	.157
.150	-.280	-.077	.073	.202	.318	-.221	-.122	-.089	-.059	.000
.250	-.237	-.107	.000	.098	.203					
.350	-.214	-.111	-.053	.027	.117					
.450	-.183	-.118	-.078	-.013	.054					
.550	-.151	-.102	-.080	-.036	.016					
.650	-.097	-.066	-.055	-.027	.000	-.031	-.020	-.053	-.092	-.105
.750	-.033	-.013	-.018	-.006	.009	-.038	.003	-.043	-.087	-.131
.850	.030	.06	.052	.043	.035	.010	.055	.016	-.036	-.112
.900	.055	.084	.071	.057	.042	.023	.078	.030	-.031	-.128

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(g)  $\alpha = 9^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>					<i>Right side</i>				
.000	.426	.880	.576	.331	-.502	.302	.676	.270	-.358	-1.146
.025	-.178	-.123	.050	-.338	-.729	.314	-.074	-.804	-1.303	-1.475
.075	-.008	-.126	-.127	-.381	-.674	.155	-.098	-.407	-1.248	-1.475
.150	.010	-.192	-.218	-.399	-.606	.077	-.095	-.272	-.607	-1.295
.250	.011	-.118	-.157	-.286	-.379	.016	-.109	-.227	-.306	-.808
.350	-.016	-.109	-.151	-.228	-.323	-.036	-.139	-.220	-.283	-.521
.450	-.050	-.127	-.156	-.203	-.300	-.074	-.150	-.195	-.245	-.417
.550	-.060	-.129	-.137	-.163	-.251	-.065	-.115	-.145	-.189	-.338
.650	-.033	-.071	-.078	-.101	-.184	-.045	-.071	-.087	-.136	-.272
.750	-.008	-.043	-.031	-.039	-.120	-.008	-.019	-.026	-.067	-.193
.850	.033	.010	.027	.034	-.020	.057	.058	.049	.013	-.076
.900	.062	.042	.064	.077	.044	.086	.096	.088	.053	-.017
	<i>Left side</i>					<i>Right side</i>				
.025	.128	-.063	-.084	.180	.530	-.657	.021	.384	.540	.639
.075	-.137	-.133	.013	.180	.495	-.349	-.042	.193	.351	.499
.150	-.200	-.118	.045	.143	.411	-.262	-.060	.110	.238	.374
.250	-.158	-.097	.039	.135	.333	-.219	-.083	.049	.152	.266
.350	-.161	-.101	.007	.111	.236	-.203	-.103	.004	.085	.190
.450	-.163	-.120	-.026	.076	.160	-.192	-.118	-.043	.023	.110
.550	-.137	-.115	-.034	.053	.117	-.148	-.101	-.051	-.001	.069
.650	-.080	-.069	-.012	.053	.099	-.085	-.063	-.025	.005	.055
.750	-.036	-.039	.009	.053	.091	-.016	.007	.018	.033	.061
.850	.033	.024	.064	.095	.119	.060	.079	.081	.080	.084
.900	.051	.042	.082	.095	.120	.088	.090	.093	.083	.072
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>					<i>Right side</i>				
.000	.229	.601	.178	-.364	-.913	.273	.531	.120	-.245	-.770
.025	.290	-.127	-.936	-1.006	-.881	.131	-.242	-.698	-.571	-.577
.075	.137	-.135	-.549	-.983	-.906	-.037	-.213	-.537	-.543	-.578
.150	.041	-.147	-.331	-.977	-.871	-.114	-.155	-.302	-.503	-.572
.250	-.028	-.146	-.261	-.851	-.837	-.065	-.088	-.169	-.431	-.542
.350	-.071	-.152	-.223	-.548	-.793					
.450	-.103	-.158	-.192	-.182	-.717					
.550	-.109	-.139	-.153	-.136	-.613					
.650	-.070	-.080	-.083	-.099	-.491	-.054	-.011	-.051	-.306	-.414
.750	-.019	-.008	-.015	-.039	-.344	-.015	.027	-.017	-.275	-.396
.850	.033	.042	.042	.017	-.225	.024	.067	.023	-.231	-.368
.900	.056	.073	.073	.050	-.167	.047	.085	.041	-.197	-.356
	<i>Left side</i>					<i>Right side</i>				
.025	-.810	-.004	.355	.515	.588	-.722	-.110	.186	.310	.384
.075	-.418	-.078	.170	.331	.466	-.386	-.179	-.029	.091	.202
.150	-.279	-.088	.084	.215	.339	-.215	-.147	-.113	-.055	.018
.250	-.248	-.124	.004	.115	.222					
.350	-.226	-.139	-.055	.031	.123					
.450	-.187	-.146	-.083	-.018	.056					
.550	-.146	-.120	-.078	-.044	.011					
.650	-.083	-.078	-.047	-.035	.005	-.024	-.013	-.034	-.070	-.113
.750	-.018	.006	-.003	.002	-.001	-.010	.015	-.022	-.078	-.137
.850	.067	.079	.070	.065	.024	.050	.080	.042	-.032	-.119
.900	.091	.100	.090	.083	.017	.067	.096	.058	-.026	-.140

TABLE VII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(h)  $\alpha = 9^\circ$ ;  $M = 0.85$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.442	.892	.583	.367	-.428	.329	.677	.338	-.212	-.920
.025	-.176	-.130	.071	-.300	-.612	.322	-.074	-.821	-1.270	-1.435
.075	-.009	-.127	-.122	-.372	-.621	.164	-.103	-.435	-1.110	-1.295
.150	.016	-.153	-.239	-.432	-.637	.081	-.098	-.283	-.683	-1.140
.250	.016	-.123	-.172	-.314	-.408	.019	-.113	-.247	-.380	-.867
.350	-.019	-.114	-.167	-.254	-.328	-.044	-.150	-.240	-.311	-.585
.450	-.054	-.140	-.176	-.222	-.332	-.084	-.163	-.216	-.270	-.469
.550	-.063	-.140	-.155	-.181	-.283	-.077	-.126	-.156	-.212	-.385
.650	-.040	-.083	-.088	-.113	-.216	-.054	-.084	-.098	-.149	-.323
.750	-.013	-.051	-.038	-.050	-.143	-.013	-.024	-.031	-.076	-.242
.850	.034	.007	.024	.032	-.053	.057	.056	.051	.012	-.115
.900	.064	.044	.059	.075	.021	.090	.099	.095	.058	-.047
	<i>Right side</i>									
.025	.143	-.058	-.107	.166	.451	-.791	-.001	.375	.594	.647
.075	-.135	-.141	-.001	.170	.466	-.371	-.063	.192	.345	.502
.150	-.216	-.126	.038	.134	.418	-.280	-.074	.103	.235	.379
.250	-.170	-.108	.034	.126	.340	-.235	-.094	.041	.154	.269
.350	-.177	-.114	.001	.101	.242	-.218	-.116	-.009	.078	.193
.450	-.175	-.131	-.041	.067	.162	-.200	-.134	-.059	.012	.109
.550	-.143	-.126	-.046	.047	.117	-.150	-.118	-.065	-.014	.061
.650	-.084	-.080	-.028	.042	.097	-.087	-.076	-.039	-.005	.044
.750	-.039	-.047	.001	.044	.084	-.013	.002	.011	.026	.050
.850	.034	.022	.055	.087	.109	.061	.079	.079	.077	.071
.900	.060	.046	.075	.094	.107	.096	.090	.091	.081	.057
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.247	.599	.240	-.285	-.804	.280	.530	.143	-.245	-.704
.025	.296	-.123	-.962	-1.136	-.847	.127	-.227	-.783	-.626	-.569
.075	.147	-.137	-.619	-1.107	-.841	-.040	-.238	-.595	-.599	-.568
.150	.046	-.151	-.351	-1.038	-.806	-.136	-.168	-.321	-.566	-.567
.250	-.026	-.157	-.274	-.813	-.767	-.074	-.097	-.167	-.485	-.552
.350	-.080	-.166	-.231	-.426	-.737					
.450	-.116	-.170	-.200	-.202	-.681					
.550	-.119	-.148	-.156	-.175	-.607					
.650	-.076	-.088	-.086	-.110	-.515	-.052	-.017	-.052	-.317	-.432
.750	-.017	-.011	-.016	-.037	-.408	-.012	.022	-.012	-.272	-.402
.850	.037	.044	.048	.025	-.303	.036	.067	.031	-.215	-.365
.900	.067	.076	.081	.057	-.248	.054	.089	.049	-.175	-.351
	<i>Right side</i>									
.025	-.903	-.023	.353	.509	.591	-.844	-.133	.182	.315	.392
.075	-.485	-.097	.166	.328	.463	-.472	-.208	-.043	.087	.206
.150	-.296	-.103	.079	.213	.338	-.220	-.168	-.137	-.084	.011
.250	-.263	-.138	-.008	.104	.225					
.350	-.235	-.156	-.068	.019	.123					
.450	-.193	-.160	-.099	-.038	.046					
.550	-.149	-.131	-.096	-.063	.000					
.650	-.079	-.083	-.059	-.045	-.019	-.034	-.016	-.036	-.081	-.126
.750	-.012	.013	-.011	-.012	-.027	-.007	.016	-.019	-.083	-.157
.850	.073	.082	.071	.055	-.003	.056	.084	.048	-.027	-.139
.900	.104	.109	.095	.074	-.014	.074	.099	.065	-.018	-.157

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(i)  $\alpha = 9^\circ$ ;  $M = 0.90$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.462	.918	.602	.408	-.211	.391	.691	.409	-.056	-.707
.025	-.189	-.123	.104	-.230	-.493	.321	-.071	-1.007	-1.368	-1.325
.075	-.005	-.125	-.106	-.315	-.560	.169	-.106	-.429	-.890	-1.188
.150	.020	-.157	-.259	-.446	-.627	.086	-.109	-.322	-.671	-1.054
.250	.017	-.129	-.191	-.357	-.516	.023	-.125	-.270	-.518	-.866
.350	-.016	-.123	-.188	-.296	-.351	-.044	-.162	-.272	-.357	-.613
.450	-.054	-.154	-.200	-.254	-.358	-.092	-.191	-.242	-.309	-.508
.550	-.067	-.157	-.177	-.197	-.335	-.087	-.147	-.168	-.238	-.433
.650	-.043	-.093	-.098	-.121	-.263	-.064	-.094	-.101	-.155	-.385
.750	-.012	-.053	-.043	-.047	-.192	-.015	-.021	-.027	-.071	-.312
.850	.041	.009	.022	.037	-.107	.063	.067	.055	.021	-.188
.900	.074	.049	.061	.088	-.041	.098	.113	.102	.070	-.113
	<i>Right side</i>									
.025	.174	-.050	-.117	.151	.292	-.842	.005	.374	.531	.643
.075	-.112	-.143	-.007	.163	.398	-.372	-.062	.190	.349	.488
.150	-.226	-.127	.035	.141	.395	-.296	-.077	.103	.238	.364
.250	-.181	-.111	.029	.119	.331	-.246	-.098	.037	.149	.255
.350	-.190	-.119	-.007	.091	.230	-.242	-.129	-.015	.080	.173
.450	-.190	-.141	-.046	.058	.143	-.220	-.151	-.069	.001	.082
.550	-.158	-.138	-.058	.033	.089	-.159	-.134	-.081	-.029	.030
.650	-.086	-.087	-.037	.032	.063	-.088	-.085	-.051	-.025	.008
.750	-.036	-.050	-.010	.036	.042	-.007	.001	.005	.015	.011
.850	.043	.025	.055	.083	.070	.075	.087	.079	.071	.029
.900	.067	.051	.076	.095	.067	.108	.106	.092	.083	.014
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.293	.614	.305	-.154	-.669	.301	.537	.174	-.257	-.629
.025	.294	-.122	-1.059	-1.412	-.831	.138	-.271	-1.054	-.817	-.584
.075	.143	-.146	-.610	-1.249	-.805	-.047	-.279	-.688	-.743	-.563
.150	.044	-.163	-.384	-.951	-.767	-.185	-.185	-.329	-.671	-.564
.250	-.029	-.177	-.333	-.699	-.727	-.075	-.098	-.122	-.596	-.563
.350	-.094	-.198	-.247	-.426	-.696					
.450	-.137	-.194	-.208	-.289	-.654					
.550	-.138	-.165	-.165	-.195	-.599					
.650	-.083	-.091	-.087	-.107	-.542	-.044	-.011	-.053	-.276	-.467
.750	-.016	-.001	-.011	-.027	-.470	-.004	.035	-.008	-.205	-.440
.850	.044	.059	.053	.039	-.385	.041	.085	.035	-.142	-.404
.900	.075	.089	.089	.072	-.335	.068	.102	.061	-.102	-.380
	<i>Right side</i>									
.025	-1.012	-.023	.346	.502	.591	-1.050	-.146	.177	.320	.391
.075	-.465	-.097	.159	.323	.450	-.561	-.257	-.057	.094	.196
.150	-.325	-.107	.073	.211	.327	-.206	-.187	-.208	-.115	-.013
.250	-.308	-.154	-.015	.102	.209					
.350	-.245	-.181	-.085	.008	.101					
.450	-.201	-.179	-.128	-.059	.014					
.550	-.151	-.146	-.120	-.087	-.046					
.650	-.076	-.103	-.071	-.062	-.065	-.028	-.006	-.038	-.080	-.158
.750	-.001	.025	-.014	-.020	-.075	.003	.027	-.016	-.078	-.210
.850	.084	.094	.068	.054	-.049	.071	.098	.057	-.015	-.186
.900	.119	.119	.095	.076	-.065	.091	.117	.073	.000	-.200

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(j)  $\alpha = 9^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.468	.919	.605	.414	-.064	.406	.687	.439	-.007	-.617
.025	-.196	-.134	.109	-.208	-.419	.316	-.069	-.985	-1.325	-1.189
.075	-.019	-.132	-.102	-.294	-.500	.159	-.112	-.445	-.865	-1.107
.150	.012	-.163	-.272	-.426	-.573	.079	-.117	-.341	-.675	-.988
.250	.010	-.137	-.205	-.382	-.543	.013	-.136	-.288	-.539	-.839
.350	-.024	-.136	-.207	-.325	-.396	-.060	-.180	-.304	-.394	-.627
.450	-.064	-.172	-.230	-.281	-.360	-.109	-.219	-.284	-.330	-.515
.550	-.081	-.180	-.204	-.214	-.353	-.105	-.171	-.182	-.255	-.457
.650	-.054	-.108	-.110	-.130	-.286	-.078	-.112	-.113	-.166	-.413
.750	-.025	-.062	-.053	-.050	-.218	-.025	-.030	-.033	-.075	-.346
.850	.031	.003	.018	.038	-.132	.052	.063	.050	.021	-.231
.900	.066	.045	.055	.086	-.066	.095	.105	.097	.072	-.149
<i>Right side</i>										
.025	.178	-.051	-.136	.145	.230	-.843	-.037	.360	.534	.653
.075	-.116	-.151	-.020	.160	.364	-.392	-.086	.176	.349	.497
.150	-.243	-.142	.025	.139	.398	-.306	-.096	.087	.237	.373
.250	-.193	-.128	.019	.119	.343	-.256	-.120	.022	.150	.263
.350	-.207	-.134	-.019	.093	.238	-.262	-.150	-.029	.076	.181
.450	-.206	-.159	-.064	.051	.147	-.244	-.180	-.090	-.004	.083
.550	-.177	-.162	-.077	.024	.088	-.168	-.163	-.103	-.036	.025
.650	-.098	-.104	-.053	.026	.057	-.095	-.103	-.073	-.036	-.002
.750	-.045	-.061	-.024	.028	.033	-.011	-.005	-.011	.003	-.005
.850	.035	.020	.039	.080	.057	.074	.081	.057	.068	.012
.900	.064	.046	.066	.091	.057	.106	.102	.085	.082	-.001
$z/b_v = 0.66$										
<i>Left side</i>										
.000	.308	.607	.327	-.104	-.609	.312	.530	.199	-.203	-.603
.025	.286	-.124	-1.076	-1.280	-.802	.126	-.243	-1.045	-.784	-.541
.075	.141	-.150	-.635	-1.109	-.764	-.049	-.262	-.953	-.728	-.539
.150	.034	-.170	-.404	-.982	-.721	-.260	-.273	-.372	-.668	-.543
.250	-.041	-.193	-.371	-.747	-.691	-.076	-.065	-.101	-.602	-.541
.350	-.114	-.235	-.328	-.525	-.661					
.450	-.159	-.219	-.201	-.357	-.627					
.550	-.159	-.185	-.171	-.234	-.584					
.650	-.097	-.096	-.093	-.120	-.538	-.052	-.015	-.056	-.293	-.471
.750	-.021	-.005	-.012	-.024	-.483	-.006	.033	-.010	-.217	-.447
.850	.042	.055	.052	.039	-.411	.042	.074	.037	-.147	-.413
.900	.074	.093	.091	.074	-.376	.066	.101	.060	-.115	-.392
<i>Right side</i>										
.025	-1.040	-.062	.329	.500	.597	-1.085	-.185	.166	.321	.406
.075	-.432	-.126	.145	.324	.458	-.655	-.296	-.068	.091	.214
.150	-.333	-.130	.062	.210	.338	-.252	-.245	-.276	-.128	-.009
.250	-.338	-.176	-.032	.097	.215					
.350	-.289	-.219	-.113	.002	.105					
.450	-.201	-.206	-.163	-.067	.015					
.550	-.156	-.164	-.143	-.102	-.052					
.650	-.077	-.113	-.085	-.073	-.075	-.041	-.014	-.043	-.083	-.167
.750	-.003	.016	-.021	-.027	-.088	-.002	.027	-.018	-.086	-.217
.850	.084	.093	.067	.053	-.066	.068	.098	.055	-.020	-.206
.900	.116	.122	.097	.076	-.081	.089	.114	.072	-.007	-.210

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(k)  $\alpha = 15.1^\circ$ ;  $M = 0.60$

$\frac{x}{C_V}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.389	.843	.692	.631	.331	-.304	.763	-.161	-1.557	-2.147
.025	-.507	-.260	.165	-.025	-.431	.375	.022	-.627	-2.092	-2.154
.075	-.195	-.216	-.051	-.166	-.440	.230	-.033	-.393	-.585	-2.343
.150	-.077	-.205	-.218	-.263	-.445	.154	-.046	-.280	-.485	-.595
.250	-.010	-.166	-.202	-.205	-.315	.084	-.062	-.234	-.407	-.512
.350	-.027	-.131	-.207	-.189	-.246	.024	-.099	-.218	-.367	-.493
.450	-.050	-.157	-.216	-.194	-.209	-.003	-.111	-.202	-.342	-.452
.550	-.066	-.145	-.195	-.161	-.158	-.006	-.085	-.166	-.309	-.385
.650	-.066	-.115	-.152	-.119	-.098	.001	-.062	-.127	-.286	-.336
.750	-.059	-.083	-.106	-.085	-.038	.024	-.017	-.067	-.224	-.271
.850	-.034	-.042	-.046	-.038	.020	.061	.036	-.007	-.131	-.214
.900	-.010	-.014	-.017	-.011	.038	.080	.064	.029	-.082	-.161
<i>Right side</i>										
.025	.257	-.131	-.416	-.038	.165	-.794	-.049	.366	.466	.507
.075	-.073	-.223	-.218	-.001	.195	-.414	-.079	.208	.337	.458
.150	-.211	-.198	-.085	.045	.214	-.308	-.074	.128	.246	.361
.250	-.209	-.161	-.017	.073	.158	-.246	-.081	.080	.168	.278
.350	-.211	-.136	-.023	.047	.149	-.225	-.099	.034	.112	.211
.450	-.223	-.159	-.049	.017	.158	-.207	-.113	-.001	.059	.144
.550	-.197	-.145	-.051	-.001	.130	-.177	-.104	-.010	.040	.110
.650	-.144	-.108	-.040	-.001	.119	-.128	-.072	-.001	.036	.096
.750	-.100	-.076	-.028	.003	.103	-.068	-.021	.027	.052	.096
.850	-.029	-.023	-.001	.031	.119	.001	.038	.073	.082	.107
.900	-.008	-.005	.009	.031	.105	.031	.050	.077	.087	.096
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	-.322	.493	-.264	-1.094	-1.140	-.025	.426	-.310	-.809	-.678
.025	.350	-.021	-.856	-1.136	-1.098	.165	-.108	-.640	-.879	-.659
.075	.209	-.053	-.601	-1.159	-1.193	-.001	-.138	-.427	-.555	-.659
.150	.105	-.067	-.333	-1.191	-1.438	-.087	-.120	-.221	-.497	-.639
.250	.036	-.079	-.223	-.582	-1.232	-.087	-.074	-.147	-.460	-.602
.350	-.010	-.090	-.191	-.150	-.717					
.450	-.043	-.099	-.168	-.170	-.334					
.550	-.059	-.081	-.145	-.166	-.228					
.650	-.036	-.051	-.088	-.117	-.156	-.057	-.014	-.074	-.286	-.588
.750	-.003	-.003	-.035	-.062	-.098	-.027	.016	-.028	-.212	-.505
.850	.027	.041	.020	-.013	-.036	.003	.048	.004	-.150	-.398
.900	.047	.061	.043	.019	-.013	.031	.064	.032	-.110	-.322
<i>Right side</i>										
.025	-.835	-.062	.350	.448	.475	-.597	-.145	.183	.246	.278
.075	-.641	-.095	.190	.316	.419	-.465	-.163	-.028	.052	.130
.150	-.325	-.081	.105	.212	.324	-.338	-.124	-.099	-.062	.010
.250	-.225	-.090	.025	.121	.218					
.350	-.202	-.104	-.019	.050	.140					
.450	-.170	-.113	-.042	-.001	.080					
.550	-.135	-.085	-.046	-.020	.043					
.650	-.087	-.046	-.028	-.015	.029	-.038	-.012	-.058	-.089	-.121
.750	-.034	-.003	-.001	-.001	.027	-.025	.011	-.028	-.092	-.117
.850	.036	.068	.055	.036	.047	.029	.064	.025	-.043	-.084
.900	.068	.091	.077	.052	.045	.047	.077	.043	-.020	-.073

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Continued

(1)  $\alpha = 15.1^\circ$ ;  $M = 0.80$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.564	.901	.802	.733	.399	.104	.826	.233	-.728	-1.492
.025	-.663	-.243	.210	.088	-.252	.374	.030	-.555	-1.504	-1.737
.075	-.195	-.218	-.014	-.094	-.343	.233	-.041	-.390	-.688	-1.840
.150	-.069	-.206	-.224	-.281	-.487	.184	-.052	-.295	-.541	-.768
.250	-.007	-.153	-.218	-.244	-.386	.096	-.066	-.247	-.449	-.551
.350	-.016	-.128	-.215	-.220	-.296	.034	-.096	-.232	-.402	-.528
.450	-.046	-.163	-.235	-.220	-.260	-.006	-.130	-.218	-.361	-.548
.550	-.057	-.153	-.209	-.178	-.207	-.006	-.095	-.175	-.327	-.445
.650	-.053	-.116	-.154	-.128	-.132	-.002	-.064	-.124	-.286	-.381
.750	-.045	-.081	-.100	-.083	-.065	.031	-.014	-.060	-.204	-.292
.850	-.021	-.032	-.035	-.025	.014	.078	.047	.011	-.095	-.215
.900	.007	.003	-.002	.008	.055	.101	.082	.052	-.034	-.155
<i>Right side</i>										
.025	.324	-.098	-.447	-.072	.138	-.729	-.052	.349	.464	.545
.075	-.022	-.227	-.223	-.009	.163	-.438	-.089	.207	.343	.462
.150	-.219	-.194	-.072	.048	.197	-.320	-.083	.140	.252	.376
.250	-.226	-.150	-.005	.083	.180	-.254	-.089	.088	.184	.295
.350	-.218	-.124	-.014	.066	.117	-.238	-.107	.041	.126	.221
.450	-.233	-.157	-.038	.028	.112	-.219	-.131	.002	.074	.154
.550	-.207	-.145	-.045	.006	.101	-.182	-.119	-.011	.048	.115
.650	-.144	-.099	-.031	.009	.091	-.127	-.078	.005	.046	.097
.750	-.089	-.070	-.014	.012	.083	-.051	-.022	.038	.066	.098
.850	-.019	-.008	.020	.041	.092	.019	.045	.088	.098	.111
.900	.005	.012	.031	.048	.084	.055	.059	.098	.109	.104
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	-.100	.535	-.029	-.793	-1.015	.004	.451	-.147	-.538	-.724
.025	.363	-.014	-.828	-1.166	-.965	.181	-.139	-.788	-.601	-.724
.075	.216	-.047	-.654	-1.178	-1.038	.002	-.160	-.716	-.571	-.727
.150	.119	-.070	-.378	-1.144	-1.169	-.103	-.136	-.240	-.527	-.716
.250	.049	-.083	-.227	-.767	-1.071	-.083	-.067	-.100	-.475	-.668
.350	-.009	-.099	-.194	-.138	-.833					
.450	-.045	-.105	-.172	-.155	-.605					
.550	-.059	-.087	-.135	-.154	-.436					
.650	-.031	-.046	-.071	-.095	-.275	-.042	.007	-.045	-.306	-.524
.750	.013	.023	-.014	-.037	-.164	-.006	.042	-.002	-.214	-.472
.850	.055	.065	.046	.021	-.065	.035	.079	.040	-.138	-.396
.900	.073	.090	.072	.045	-.028	.064	.102	.066	-.088	-.339
<i>Right side</i>										
.025	-.848	-.040	.375	.465	.505	-.896	-.142	.201	.281	.300
.075	-.690	-.078	.206	.329	.427	-.371	-.177	-.017	.080	.155
.150	-.382	-.067	.118	.237	.339	-.318	-.130	-.100	-.049	.006
.250	-.226	-.089	.051	.137	.235					
.350	-.198	-.107	-.014	.057	.147					
.450	-.163	-.101	-.040	.009	.081					
.550	-.124	-.080	-.043	-.015	.038					
.650	-.068	-.037	-.022	-.009	.018	-.007	.013	-.011	-.088	-.137
.750	-.009	.018	.015	.008	.015	.008	.042	-.002	-.083	-.143
.850	.070	.093	.078	.058	.038	.064	.103	.058	-.026	-.106
.900	.101	.119	.106	.074	.038	.086	.116	.071	-.002	-.108

TABLE VII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE AND VERTICAL

TAIL - Concluded

(m)  $\alpha = 15.10^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.599	.924	.838	.749	.401	.192	.843	.344	-.566	-1.224
.025	-.714	-.223	.214	.113	-.180	.355	.018	-.513	-1.356	-1.597
.075	-.216	-.219	-.005	-.076	-.276	.216	-.052	-.407	-.722	-1.630
.150	-.082	-.210	-.231	-.287	-.461	.153	-.063	-.315	-.593	-.826
.250	-.016	-.159	-.239	-.278	-.444	.085	-.078	-.266	-.492	-.590
.350	-.032	-.133	-.233	-.244	-.331	.024	-.110	-.256	-.431	-.511
.450	-.059	-.176	-.263	-.248	-.283	-.019	-.146	-.239	-.387	-.649
.550	-.069	-.159	-.226	-.198	-.243	-.016	-.110	-.190	-.344	-.476
.650	-.062	-.123	-.167	-.139	-.173	-.011	-.076	-.139	-.301	-.425
.750	-.051	-.083	-.110	-.097	-.095	.019	-.021	-.062	-.212	-.321
.850	-.025	-.028	-.039	-.028	-.002	.068	.048	.008	-.100	-.230
.900	-.001	.009	-.005	.002	.044	.097	.085	.056	-.038	-.167
<i>Right side</i>										
.025	.338	-.096	-.498	-.089	.116	-.722	-.058	.333	.455	.546
.075	-.014	-.241	-.233	-.025	.140	-.471	-.096	.199	.337	.461
.150	-.234	-.190	-.078	.041	.182	-.346	-.088	.136	.252	.372
.250	-.249	-.136	-.009	.080	.170	-.279	-.090	.091	.183	.291
.350	-.237	-.125	-.021	.057	.110	-.257	-.116	.038	.123	.219
.450	-.252	-.159	-.051	.019	.085	-.237	-.139	-.008	.068	.149
.550	-.224	-.149	-.058	-.001	.078	-.194	-.123	-.021	.041	.108
.650	-.154	-.099	-.039	-.001	.064	-.133	-.085	-.006	.036	.084
.750	-.104	-.068	-.025	-.002	.058	-.059	-.026	.036	.061	.088
.850	-.026	-.008	.014	.035	.077	.024	.048	.086	.094	.096
.900	.002	.016	.031	.038	.068	.054	.061	.102	.101	.090
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	-.061	.546	.032	-.686	-1.031	.032	.454	-.115	-.544	-.728
.025	.355	-.025	-.882	-1.209	-.962	.170	-.146	-.960	-.636	-.731
.075	.212	-.058	-.706	-1.221	-.989	-.014	-.184	-.770	-.599	-.708
.150	.108	-.076	-.395	-1.172	-1.016	-.137	-.147	-.290	-.558	-.695
.250	.037	-.093	-.250	-.794	-.966	-.110	-.075	-.096	-.505	-.668
.350	-.022	-.109	-.213	-.172	-.831					
.450	-.065	-.116	-.182	-.195	-.675					
.550	-.081	-.093	-.146	-.185	-.517					
.650	-.045	-.049	-.082	-.116	-.354	-.049	.014	-.051	-.318	-.519
.750	.005	.028	-.015	-.051	-.239	-.011	.045	.001	-.231	-.465
.850	.048	.073	.048	.013	-.127	.035	.092	.051	-.149	-.398
.900	.070	.096	.078	.039	-.080	.065	.112	.078	-.103	-.349
<i>Right side</i>										
.025	-.952	-.036	.375	.460	.506	-.859	-.139	.199	.285	.308
.075	-.794	-.080	.204	.325	.428	-.509	-.196	-.033	.078	.157
.150	-.411	-.070	.122	.232	.337	-.349	-.145	-.125	-.066	.004
.250	-.237	-.093	.044	.136	.230					
.350	-.214	-.110	-.021	.049	.137					
.450	-.181	-.113	-.053	-.004	.070					
.550	-.137	-.088	-.056	-.034	.021					
.650	-.080	-.042	-.029	-.027	.001	-.028	.014	-.015	-.109	-.168
.750	-.014	.022	.015	-.002	-.006	.002	.051	.001	-.097	-.174
.850	.067	.105	.088	.051	.019	.063	.115	.085	-.037	-.140
.900	.100	.130	.106	.064	.018	.080	.129	.085	-.014	-.141



TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$

(a)  $\alpha = -3^\circ$ ;  $M = 0.60$

$\frac{x}{C_V}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.683	.825	.474	-.359	-1.024	.407	.876	.499	-.132	-.785
.025	.264	-.145	-.671	-1.258	-1.373	.365	-.115	-.720	-1.159	-1.107
.075	.103	-.170	-.457	-.766	-1.316	.135	-.186	-.552	-1.049	-1.143
.150	-.017	-.214	-.411	-.587	-1.118	.024	-.207	-.439	-.748	-.939
.250	-.056	-.195	-.317	-.398	-.833	-.070	-.234	-.404	-.500	-.909
.350	-.082	-.184	-.275	-.300	-.587	-.139	-.267	-.404	-.410	-.766
.450	-.132	-.207	-.271	-.265	-.417	-.181	-.280	-.386	-.352	-.596
.550	-.128	-.184	-.229	-.215	-.302	-.176	-.248	-.319	-.297	-.454
.650	-.093	-.122	-.162	-.148	-.215	-.144	-.193	-.241	-.251	-.352
.750	-.047	-.076	-.095	-.097	-.159	-.095	-.120	-.151	-.196	-.288
.850	-.001	-.012	-.019	-.031	-.104	-.008	-.028	-.049	-.129	-.244
.900	.029	.029	.027	.008	-.067	.020	.011	.001	-.086	-.208
	<i>Right side</i>									
.025	-.579	-.101	.203	.590	.818	-.985	-.129	.386	.641	.811
.075	-.499	-.198	.077	.305	.526	-.568	-.202	.133	.349	.542
.150	-.416	-.212	-.017	.153	.339	-.464	-.218	.013	.181	.362
.250	-.333	-.202	-.056	.073	.225	-.402	-.239	-.075	.064	.215
.350	-.296	-.195	-.095	-.001	.119	-.393	-.262	-.137	-.031	.105
.450	-.287	-.216	-.135	-.067	.034	-.370	-.278	-.188	-.097	.015
.550	-.236	-.193	-.128	-.086	-.001	-.314	-.255	-.171	-.123	-.033
.650	-.160	-.131	-.089	-.063	-.010	-.234	-.191	-.142	-.118	-.056
.750	-.091	-.074	-.040	-.035	-.005	-.137	-.111	-.079	-.081	-.056
.850	-.006	-.001	.017	.013	.015	-.031	-.017	.001	-.028	-.040
.900	.022	.034	.041	.031	.018	.006	.011	.024	-.019	-.067
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.379	.876	.462	-.024	-.428	.428	.743	.252	.199	-.274
.025	.393	-.177	-1.135	-1.086	-.833	.296	-.521	-1.669	-.440	-.403
.075	.149	-.239	-.681	-1.065	-.856	-.006	-.588	-1.347	-.438	-.405
.150	-.010	-.292	-.600	-.861	-.881	-.250	-.627	-1.324	-.426	-.412
.250	-.125	-.340	-.554	-.532	-.838	-.324	-.544	-1.093	-.433	-.426
.350	-.208	-.377	-.547	-.392	-.686					
.450	-.264	-.390	-.515	-.371	-.502					
.550	-.261	-.349	-.432	-.375	-.488					
.650	-.208	-.248	-.314	-.398	-.516	-.245	-.280	-.294	-.470	-.481
.750	-.137	-.161	-.204	-.435	-.532	-.146	-.152	-.185	-.470	-.477
.850	-.056	-.076	-.098	-.424	-.513	-.056	-.040	-.091	-.438	-.435
.900	-.031	-.030	-.047	-.405	-.511	-.015	.006	-.042	-.410	-.417
	<i>Right side</i>									
.025	-1.026	-.166	.418	.657	.802	-1.586	-.487	.326	.553	.799
.075	-.711	-.260	.133	.351	.542	-1.328	-.615	-.031	.188	.461
.150	-.579	-.283	.001	.179	.362	-1.365	-.601	-.231	-.054	.197
.250	-.570	-.345	-.123	.018	.197					
.350	-.568	-.381	-.206	-.100	.064					
.450	-.515	-.381	-.252	-.180	-.042					
.550	-.427	-.342	-.248	-.208	-.104					
.650	-.317	-.255	-.197	-.196	-.129	-.277	-.239	-.190	-.238	-.182
.750	-.199	-.168	-.123	-.173	-.146	-.183	-.143	-.128	-.221	-.215
.850	-.072	-.044	-.024	-.123	-.136	-.077	-.017	-.015	-.171	-.182
.900	-.026	-.005	.004	-.116	-.173	-.031	.032	.017	-.187	-.201

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(b)  $\alpha = -9^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.840	.891	.697	.282	-.215	.678	.919	.759	.374	-.085
.025	.276	-.110	-.619	-1.099	-.963	.362	-.087	-1.129	-1.612	-.638
.075	.117	-.149	-.448	-.955	-.860	.141	-.182	-.469	-1.433	-.644
.150	-.010	-.221	-.428	-.830	-.790	.021	-.216	-.423	-1.258	-.666
.250	-.063	-.218	-.314	-.372	-.674	-.083	-.262	-.380	-.317	-.695
.350	-.100	-.213	-.267	-.247	-.600	-.189	-.328	-.389	-.293	-.664
.450	-.177	-.270	-.282	-.236	-.534	-.273	-.389	-.380	-.302	-.588
.550	-.200	-.257	-.236	-.203	-.461	-.291	-.362	-.313	-.264	-.490
.650	-.155	-.164	-.156	-.143	-.389	-.253	-.279	-.239	-.233	-.417
.750	-.091	-.096	-.084	-.097	-.313	-.148	-.142	-.158	-.193	-.343
.850	-.015	-.009	-.006	-.034	-.232	-.048	-.029	-.056	-.140	-.288
.900	.027	.036	.043	.002	-.190	.004	.026	.000	-.097	-.253
	<i>Right side</i>									
.025	-.492	-.053	.319	.599	.830	-1.076	-.109	.378	.640	.819
.075	-.495	-.190	.092	.320	.542	-.530	-.201	.132	.359	.560
.150	-.423	-.219	-.009	.175	.360	-.427	-.227	.013	.200	.380
.250	-.326	-.221	-.066	.083	.239	-.371	-.267	-.083	.070	.230
.350	-.281	-.230	-.115	-.005	.125	-.366	-.328	-.179	-.047	.105
.450	-.279	-.273	-.187	-.094	.022	-.362	-.383	-.262	-.149	-.008
.550	-.235	-.260	-.202	-.138	-.041	-.310	-.359	-.279	-.207	-.085
.650	-.149	-.168	-.145	-.112	-.050	-.230	-.248	-.234	-.196	-.117
.750	-.074	-.087	-.087	-.086	-.057	-.138	-.127	-.132	-.143	-.111
.850	.010	.008	.002	-.021	-.044	-.039	-.010	-.026	-.073	-.099
.900	.047	.040	.029	.002	-.054	.005	.029	-.001	-.059	-.126
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.698	.936	.773	.403	.008	.722	.825	.684	.392	-.097
.025	.378	-.124	-1.214	-1.634	-.727	.239	-.418	-1.201	-.377	-.398
.075	.151	-.210	-.808	-1.400	-.741	-.054	-.570	-.521	-.360	-.385
.150	-.015	-.293	-.509	-.901	-.761	-.346	-.799	-.480	-.346	-.381
.250	-.160	-.389	-.445	-.591	-.706	-.363	-.716	-.474	-.351	-.386
.350	-.294	-.515	-.445	-.342	-.658					
.450	-.432	-.612	-.438	-.297	-.553					
.550	-.482	-.540	-.389	-.322	-.455					
.650	-.385	-.253	-.331	-.363	-.458	-.351	-.294	-.426	-.388	-.429
.750	-.193	-.173	-.271	-.398	-.472	-.242	-.156	-.414	-.394	-.423
.850	-.128	-.060	-.224	-.420	-.466	-.175	-.043	-.374	-.377	-.401
.900	-.106	-.014	-.193	-.430	-.472	-.154	.009	-.339	-.369	-.389
	<i>Right side</i>									
.025	-1.196	-.118	.407	.667	.825	-1.100	-.380	.255	.591	.837
.075	-.799	-.237	.131	.375	.571	-.440	-.685	-.099	.218	.504
.150	-.472	-.285	-.007	.198	.391	-.418	-.765	-.331	-.050	.218
.250	-.433	-.405	-.168	.025	.216					
.350	-.432	-.537	-.299	-.132	.059					
.450	-.409	-.589	-.418	-.268	-.082					
.550	-.359	-.512	-.443	-.342	-.175					
.650	-.308	-.305	-.360	-.322	-.210	-.397	-.259	-.322	-.346	-.261
.750	-.274	-.170	-.179	-.252	-.200	-.395	-.149	-.216	-.270	-.245
.850	-.230	-.029	-.090	-.144	-.160	-.368	-.026	-.121	-.181	-.178
.900	-.198	.025	-.069	-.167	-.187	-.337	.025	-.116	-.183	-.189

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(c)  $\alpha = -9^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.877	.914	.755	-.391	-.063	.734	.938	-1.022	-1.492	-.074
.025	.286	-.092	-.556	-1.022	-.771	.363	-.075	-1.022	-1.415	-.598
.075	.129	-.137	-.503	-.885	-.706	.144	-.173	-.788	-1.284	-.608
.150	-.008	-.219	-.437	-.833	-.681	.027	-.209	-.427	-1.193	-.628
.250	-.060	-.217	-.315	-.688	-.618	-.080	-.251	-.368	-.816	-.650
.350	-.106	-.213	-.269	-.251	-.586	-.187	-.325	-.383	-.178	-.636
.450	-.197	-.296	-.289	-.199	-.563	-.297	-.421	-.399	-.212	-.588
.550	-.253	-.321	-.249	-.179	-.506	-.354	-.464	-.328	-.237	-.517
.650	-.214	-.213	-.161	-.142	-.448	-.371	-.407	-.246	-.231	-.454
.750	-.151	-.112	-.088	-.103	-.383	-.276	-.125	-.162	-.205	-.396
.850	-.040	-.008	-.008	-.045	-.307	-.067	-.016	-.065	-.159	-.353
.900	.012	.046	.042	-.005	-.274	-.011	.041	-.007	-.117	-.316
	<i>Right side</i>									
.025	-.446	-.025	.329	.600	.827	-.997	-.076	.379	.637	.816
.075	-.578	-.172	.102	.322	.543	-.793	-.184	.135	.360	.557
.150	-.468	-.211	-.004	.170	.357	-.426	-.213	.018	.205	.380
.250	-.333	-.216	-.065	.081	.236	-.354	-.249	-.087	.074	.232
.350	-.284	-.230	-.121	-.010	.123	-.357	-.324	-.185	-.050	.097
.450	-.287	-.290	-.208	-.116	.004	-.371	-.417	-.296	-.175	-.030
.550	-.243	-.313	-.256	-.175	-.067	-.316	-.474	-.360	-.255	-.124
.650	-.150	-.203	-.206	-.156	-.090	-.233	-.307	-.359	-.274	-.170
.750	-.078	-.099	-.136	-.133	-.113	-.141	-.122	-.198	-.211	-.173
.850	.009	.011	-.025	-.053	-.094	-.053	-.001	-.059	-.104	-.146
.900	.047	.053	.016	-.021	-.113	-.010	.045	-.015	-.086	-.174
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.752	.958	.829	.509	.122	.770	.861	.722	.419	.026
.025	.390	-.098	-1.101	-1.469	-.694	.250	-.345	-1.007	-.879	-.416
.075	.167	-.184	-.905	-1.310	-.698	-.018	-.488	-.447	-.363	-.398
.150	.004	-.264	-.596	-1.179	-.706	-.330	-.733	-.406	-.349	-.390
.250	-.143	-.354	-.368	-.481	-.660	-.320	-.730	-.402	-.350	-.397
.350	-.286	-.482	-.399	-.347	-.590					
.450	-.431	-.622	-.439	-.344	-.530					
.550	-.561	-.724	-.385	-.320	-.468					
.650	-.633	-.639	-.326	-.349	-.473	-.858	-.418	-.407	-.380	-.426
.750	-.511	-.085	-.291	-.386	-.483	-.370	-.281	-.405	-.385	-.423
.850	-.097	-.043	-.259	-.409	-.473	-.211	-.123	-.375	-.374	-.404
.900	-.108	-.005	-.241	-.426	-.484	-.181	-.059	-.356	-.363	-.396
	<i>Right side</i>									
.025	-1.078	-.076	.411	.667	.826	-.991	-.267	.280	.614	.843
.075	-.900	-.204	.142	.380	.573	-.386	-.580	-.074	.248	.510
.150	-.651	-.249	.008	.209	.393	-.363	-.692	-.312	-.031	.220
.250	-.337	-.361	-.151	.035	.214					
.350	-.394	-.495	-.298	-.127	.049					
.450	-.404	-.629	-.439	-.285	-.106					
.550	-.354	-.716	-.566	-.395	-.230					
.650	-.304	-.529	-.597	-.468	-.311	-.370	-.375	-.609	-.660	-.480
.750	-.291	-.119	-.420	-.429	-.310	-.378	-.273	-.463	-.393	-.290
.850	-.273	-.019	-.074	-.135	-.181	-.361	-.139	-.269	-.212	-.187
.900	-.250	.024	-.082	-.165	-.204	-.340	-.070	-.205	-.185	-.187

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(d)  $\alpha = -9^\circ$ ;  $M = 0.90$

$\frac{x}{C_V}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.915	.937	.809	.497	.057	.794	.954	.865	.597	.167
.025	.303	-.071	-.482	-.921	-.821	.374	-.054	-.898	-1.209	-.613
.075	.147	-.119	-.483	-.785	-.736	.162	-.161	-.745	-1.117	-.618
.150	.008	-.211	-.449	-.770	-.702	.046	-.200	-.563	-1.055	-.647
.250	-.051	-.212	-.490	-.776	-.636	-.056	-.234	-.521	-.979	-.664
.350	-.089	-.199	-.264	-.520	-.578	-.169	-.310	-.358	-.674	-.640
.450	-.196	-.299	-.291	-.203	-.543	-.283	-.395	-.324	-.101	-.581
.550	-.308	-.377	-.252	-.120	-.486	-.375	-.493	-.345	-.147	-.502
.650	-.264	-.304	-.167	-.108	-.425	-.429	-.547	-.259	-.201	-.441
.750	-.312	-.173	-.098	-.099	-.363	-.449	-.393	-.172	-.214	-.409
.850	-.134	-.009	-.012	-.056	-.295	-.386	.029	-.081	-.195	-.390
.900	-.024	.048	.037	-.022	-.266	-.075	.068	-.020	-.153	-.348
	<i>Right side</i>									
.025	-.365	-.002	.347	.615	.841	-.859	-.064	.383	.641	.829
.075	-.518	-.156	.113	.340	.557	-.754	-.177	.148	.372	.575
.150	-.459	-.207	.003	.187	.374	-.607	-.203	.030	.217	.398
.250	-.503	-.214	-.056	.095	.252	-.492	-.234	-.066	.083	.249
.350	-.288	-.215	-.110	-.001	.134	-.334	-.307	-.171	-.039	.114
.450	-.268	-.277	-.198	-.110	.009	-.284	-.390	-.286	-.170	-.024
.550	-.235	-.401	-.325	-.206	-.079	-.319	-.508	-.395	-.263	-.129
.650	-.148	-.292	-.267	-.210	-.117	-.239	-.571	-.433	-.341	-.203
.750	-.082	-.139	-.283	-.260	-.174	-.152	-.157	-.452	-.376	-.252
.850	.007	.012	-.087	-.149	-.168	-.066	.020	-.191	-.293	-.256
.900	.050	.056	-.018	-.087	-.189	-.024	.059	-.072	-.228	-.289
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.800	.977	.878	.602	.206	.813	.893	.784	.561	.124
.025	.408	-.067	-.979	-1.263	-.731	.304	-.254	-.872	-.856	-.402
.075	.190	-.160	-.829	-1.139	-.737	.036	-.419	-.401	-.347	-.382
.150	.036	-.235	-.763	-1.062	-.725	-.281	-.664	-.356	-.340	-.378
.250	-.106	-.322	-.652	-.848	-.640	-.307	-.687	-.351	-.340	-.378
.350	-.253	-.440	-.151	-.405	-.502					
.450	-.400	-.571	-.336	-.349	-.479					
.550	-.521	-.702	-.394	-.344	-.440					
.650	-.638	-.705	-.322	-.360	-.447	-.881	-.447	-.378	-.367	-.402
.750	-.677	-.511	-.325	-.386	-.455	-.846	-.431	-.386	-.372	-.401
.850	-.626	-.010	-.318	-.405	-.447	-.554	-.330	-.368	-.360	-.386
.900	-.515	.022	-.306	-.422	-.455	-.412	-.260	-.355	-.353	-.381
	<i>Right side</i>									
.025	-.942	-.053	.424	.672	.845	-.859	-.195	.318	.633	.863
.075	-.807	-.181	.167	.395	.592	-.351	-.496	-.035	.279	.534
.150	-.725	-.218	.033	.230	.414	-.322	-.625	-.278	.007	.251
.250	-.654	-.323	-.121	.061	.244					
.350	-.136	-.453	-.270	-.102	.072					
.450	-.300	-.588	-.401	-.260	-.089					
.550	-.339	-.687	-.552	-.405	-.220					
.650	-.289	-.689	-.639	-.482	-.320	-.335	-.411	-.779	-.741	-.564
.750	-.313	-.353	-.695	-.562	-.426	-.347	-.372	-.779	-.783	-.631
.850	-.319	.017	-.620	-.572	-.443	-.339	-.284	-.508	-.610	-.459
.900	-.311	.037	-.340	-.564	-.436	-.326	-.226	-.421	-.429	-.279

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(c)  $\alpha = -9^\circ$ ;  $M = 0.92$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.931	.950	.879	.522	.100	.809	.263	.883	.621	.204
.025	.300	-.061	-.452	-.856	-.809	.367	-.041	-.849	-1.170	-.604
.075	.147	-.113	-.464	-.758	-.718	.159	-.160	-.715	-1.084	-.611
.150	.006	-.209	-.454	-.744	-.686	.043	-.202	-.579	-1.021	-.628
.250	-.051	-.222	-.520	-.756	-.624	-.055	-.227	-.503	-.956	-.648
.350	-.095	-.199	-.328	-.576	-.567	-.167	-.304	-.489	-.833	-.633
.450	-.199	-.301	-.311	-.234	-.534	-.279	-.374	-.334	-.122	-.586
.550	-.308	-.373	-.234	-.104	-.491	-.381	-.485	-.262	-.100	-.511
.650	-.271	-.357	-.164	-.086	-.441	-.456	-.567	-.283	-.168	-.454
.750	-.347	-.350	-.106	-.086	-.389	-.458	-.534	-.196	-.197	-.418
.850	-.228	-.043	-.020	-.056	-.330	-.482	-.057	-.101	-.186	-.396
.900	-.074	.037	.030	-.020	-.291	-.372	.081	-.040	-.154	-.359
<i>Right side</i>										
.025	-.348	.003	.351	.623	.851	-.824	-.067	.389	.646	.839
.075	-.517	-.158	.121	.348	.566	-.751	-.186	.155	.377	.586
.150	-.441	-.213	.013	.196	.385	-.619	-.211	.038	.227	.408
.250	-.543	-.230	-.048	.103	.267	-.554	-.235	-.057	.095	.260
.350	-.345	-.214	-.103	.008	.147	-.462	-.307	-.162	-.026	.125
.450	-.300	-.303	-.189	-.102	.019	-.281	-.393	-.270	-.158	-.014
.550	-.220	-.411	-.336	-.196	-.077	-.263	-.492	-.391	-.249	-.120
.650	-.154	-.340	-.271	-.198	-.112	-.256	-.587	-.460	-.329	-.195
.750	-.095	-.291	-.340	-.275	-.188	-.178	-.494	-.445	-.371	-.256
.850	-.006	-.012	-.175	-.174	-.188	-.098	-.017	-.433	-.333	-.262
.900	.030	.047	-.058	-.108	-.250	-.051	.053	-.209	-.306	-.324
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.818	.986	.899	.625	.240	.825	.908	.803	.575	.118
.025	.405	-.053	-.924	-1.226	-.707	.312	-.218	-.826	-.921	-.404
.075	.193	-.150	-.781	-1.102	-.707	.047	-.382	-.391	-.331	-.385
.150	.038	-.227	-.734	-1.042	-.709	-.267	-.635	-.344	-.329	-.378
.250	-.102	-.313	-.658	-.925	-.647	-.309	-.664	-.338	-.330	-.378
.350	-.247	-.424	-.416	-.426	-.538					
.450	-.392	-.551	-.228	-.331	-.504					
.550	-.514	-.684	-.330	-.329	-.441					
.650	-.638	-.744	-.336	-.347	-.439	-.895	-.480	-.371	-.357	-.401
.750	-.698	-.691	-.351	-.371	-.446	-.840	-.461	-.380	-.358	-.402
.850	-.660	-.230	-.351	-.387	-.439	-.594	-.422	-.370	-.350	-.393
.900	-.664	-.031	-.344	-.413	-.450	-.494	-.371	-.360	-.355	-.384
<i>Right side</i>										
.025	-.905	-.051	.433	.679	.854	-.835	-.174	.338	.643	.875
.075	-.789	-.182	.179	.404	.606	-.364	-.469	-.007	.292	.553
.150	-.718	-.219	.049	.244	.429	-.331	-.613	-.258	.021	.274
.250	-.686	-.316	-.102	.075	.257					
.350	-.417	-.439	-.254	-.086	.089					
.450	-.208	-.572	-.380	-.244	-.069					
.550	-.308	-.687	-.527	-.385	-.204					
.650	-.313	-.734	-.622	-.469	-.305	-.355	-.476	-.770	-.724	-.548
.750	-.347	-.628	-.698	-.547	-.412	-.363	-.445	-.801	-.784	-.611
.850	-.356	-.098	-.665	-.559	-.435	-.355	-.379	-.572	-.658	-.565
.900	-.364	.014	-.642	-.587	-.466	-.348	-.324	-.475	-.522	-.463

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(f)  $\alpha = 0^\circ$ ;  $M = 0.60$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.615	.640	.311	-.441	-1.142	.245	.714	-.297	-.420	-.973
.025	.266	-.103	-.349	-1.088	-1.760	.336	-.085	-.649	-1.034	-.898
.075	.132	-.110	-.368	-.647	-1.344	.159	-.123	-.449	-.907	-.875
.150	.041	-.137	-.309	-.459	-.827	.066	-.132	-.336	-.666	-.820
.250	.021	-.098	-.214	-.286	-.398	-.002	-.146	-.289	-.400	-.767
.350	-.024	-.110	-.187	-.222	-.307	-.058	-.166	-.284	-.304	-.669
.450	-.056	-.128	-.178	-.200	-.273	-.083	-.182	-.262	-.259	-.551
.550	-.058	-.107	-.137	-.150	-.262	-.083	-.157	-.212	-.213	-.444
.650	-.033	-.069	-.087	-.107	-.187	-.063	-.119	-.151	-.172	-.364
.750	-.011	-.031	-.033	-.068	-.122	-.029	-.060	-.078	-.118	-.278
.850	.030	.028	.023	-.016	-.049	.030	.019	.005	-.047	-.198
.900	.048	.048	.051	.012	-.002	.057	.055	.051	-.011	-.135
<i>Right side</i>										
.025	-.462	-.028	.315	.584	.820	-.643	-.067	.381	.600	.749
.075	-.389	-.107	.127	.339	.551	-.430	-.121	.166	.371	.547
.150	-.299	-.119	.055	.216	.396	-.335	-.132	.071	.232	.405
.250	-.222	-.105	.023	.143	.296	-.281	-.144	.005	.134	.287
.350	-.194	-.121	-.022	.073	.200	-.267	-.164	-.047	.064	.205
.450	-.183	-.132	-.063	.014	.120	-.253	-.177	-.078	.005	.125
.550	-.138	-.134	-.063	-.007	.089	-.210	-.159	-.083	-.022	.076
.650	-.088	-.076	-.033	-.002	.080	-.145	-.119	-.065	-.016	.058
.750	-.031	-.028	-.011	.016	.073	-.072	-.065	-.020	.000	.049
.850	.035	.030	.035	.048	.089	.016	.024	.041	.039	.058
.900	.050	.053	.051	.053	.089	.035	.044	.053	.046	.040
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.198	.721	.207	-.295	-.533	.393	.775	.098	-.227	-.727
.025	.379	-.110	-.877	-.952	-.618	.484	-.252	-1.185	-.852	-.538
.075	.202	-.137	-.535	-.904	-.633	.205	-.315	-.988	-.343	-.362
.150	.089	-.157	-.425	-.813	-.651	-.013	-.333	-1.013	-.336	-.358
.250	.007	-.175	-.377	-.588	-.615	-.092	-.304	-.818	-.241	-.367
.350	-.063	-.211	-.368	-.372	-.560					
.450	-.106	-.227	-.341	-.272	-.480					
.550	-.120	-.216	-.280	-.261	-.407					
.650	-.092	-.153	-.198	-.263	-.382	-.108	-.159	-.171	-.382	-.402
.750	-.047	-.080	-.106	-.272	-.362	-.056	-.074	-.085	-.377	-.371
.850	.010	-.006	-.020	-.247	-.327	.012	.014	.005	-.347	-.313
.900	.025	.026	.014	-.229	-.304	.035	.053	.041	-.316	-.284
<i>Right side</i>										
.025	-.823	-.055	.426	.630	.736	-1.183	-.166	.526	.723	.909
.075	-.519	-.125	.207	.403	.569	-1.018	-.290	.200	.403	.640
.150	-.385	-.134	.105	.266	.433	-1.011	-.288	.017	.199	.418
.250	-.367	-.177	.019	.150	.316					
.350	-.369	-.207	-.054	.062	.213					
.450	-.326	-.227	-.097	-.009	.131					
.550	-.276	-.209	-.115	-.050	.067					
.650	-.192	-.162	-.087	-.057	.029	-.165	-.155	-.097	-.104	-.024
.750	-.106	-.078	-.045	-.043	.002	-.083	-.078	-.078	-.122	-.073
.850	-.002	.010	.028	-.011	-.011	.019	.024	.026	-.088	-.062
.900	.035	.042	.035	-.020	-.047	.055	.069	.055	-.113	-.093

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(g)  $\alpha = 0^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.741	.668	.514	.181	-.215	.480	.775	.549	.043	-.396
.025	.270	-.068	-.534	-1.143	-.895	.337	-.068	-.014	-1.583	-.620
.075	.140	-.095	-.363	-.846	-.802	.158	-.122	-.431	-1.120	-.619
.150	.045	-.131	-.315	-.486	-.731	.069	-.137	-.326	-.725	-.626
.250	.022	-.101	-.211	-.282	-.622	.000	-.155	-.273	-.383	-.628
.350	-.023	-.121	-.179	-.227	-.529	-.067	-.190	-.270	-.283	-.594
.450	-.070	-.146	-.182	-.199	-.465	-.123	-.217	-.265	-.236	-.534
.550	-.079	-.128	-.142	-.149	-.392	-.124	-.195	-.211	-.190	-.465
.650	-.055	-.083	-.083	-.106	-.314	-.100	-.143	-.154	-.158	-.408
.750	-.024	-.033	-.026	-.058	-.244	-.049	-.060	-.075	-.117	-.339
.850	.030	.030	.032	-.010	-.151	.024	.029	.014	-.047	-.282
.900	.056	.068	.069	.022	-.103	.065	.074	.063	-.003	-.242
<i>Right side</i>										
.025	-.397	.003	.324	.591	.815	-.916	-.045	.393	.611	.749
.075	-.384	-.095	.142	.357	.556	-.419	-.109	.175	.378	.545
.150	-.299	-.110	.068	.233	.397	-.323	-.125	.080	.248	.400
.250	-.219	-.101	.030	.161	.295	-.272	-.146	.009	.146	.280
.350	-.195	-.124	-.023	.077	.189	-.265	-.176	-.050	.066	.192
.450	-.189	-.145	-.071	.010	.101	-.263	-.204	-.103	-.005	.095
.550	-.147	-.131	-.081	-.017	.052	-.224	-.188	-.122	-.049	.034
.650	-.083	-.077	-.053	-.014	.032	-.153	-.137	-.092	-.050	.010
.750	-.023	-.029	-.023	-.005	.020	-.061	-.048	-.035	-.028	-.006
.850	.042	.041	.039	.034	.025	.030	.047	.038	.016	-.015
.900	.074	.066	.057	.046	.017	.063	.065	.056	.018	-.048
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.467	.783	.508	.093	-.285	.666	.837	.561	.178	-.376
.025	.380	-.084	-1.093	-.822	-.619	.430	-.277	-.864	-.828	-.432
.075	.204	-.122	-.486	-.802	-.634	.161	-.401	-.409	-.297	-.382
.150	.087	-.160	-.363	-.719	-.652	-.068	-.413	-.406	-.291	-.371
.250	.000	-.204	-.300	-.545	-.644	-.142	-.357	-.401	-.298	-.383
.350	-.091	-.265	-.303	-.359	-.616					
.450	-.166	-.302	-.308	-.267	-.520					
.550	-.194	-.274	-.280	-.252	-.440					
.650	-.151	-.185	-.214	-.261	-.423	-.168	-.160	-.309	-.344	-.420
.750	-.076	-.077	-.143	-.271	-.417	-.107	-.047	-.283	-.351	-.405
.850	-.008	.009	-.071	-.264	-.396	-.042	.057	-.234	-.336	-.367
.900	.021	.053	-.030	-.250	-.380	-.021	.103	-.198	-.321	-.344
<i>Right side</i>										
.025	-1.046	-.030	.433	.641	.752	-.935	-.139	.490	.757	.941
.075	-.459	-.109	.211	.410	.576	-.552	-.374	.155	.429	.655
.150	-.340	-.130	.112	.285	.442	-.542	-.344	-.026	.195	.407
.250	-.333	-.195	.006	.164	.316					
.350	-.352	-.252	-.084	.052	.195					
.450	-.340	-.287	-.155	-.041	.090					
.550	-.287	-.267	-.187	-.102	.005					
.650	-.204	-.208	-.151	-.114	-.043	-.262	-.160	-.172	-.170	-.109
.750	-.123	-.057	-.087	-.100	-.072	-.225	-.060	-.130	-.174	-.147
.850	-.018	.044	.002	-.056	-.072	-.160	.057	-.041	-.120	-.121
.900	.027	.075	.030	-.067	-.112	-.123	.097	-.039	-.141	-.154

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(h)  $\alpha = 0^\circ$ ;  $M = 0.85$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.774	-.676	-.563	-.301	-.058	.536	-.784	-.607	-.199	-.226
.025	.272	-.069	-.527	-1.057	-.834	.336	-.078	-1.028	-1.470	-.558
.075	.146	-.100	-.375	-.827	-.719	.158	-.136	-.410	-1.308	-.558
.150	.046	-.146	-.336	-.655	-.653	.070	-.152	-.330	-.864	-.567
.250	.019	-.117	-.222	-.293	-.570	-.007	-.173	-.278	-.254	-.572
.350	-.029	-.137	-.192	-.213	-.509	-.077	-.214	-.275	-.287	-.556
.450	-.089	-.173	-.194	-.200	-.478	-.145	-.258	-.276	-.256	-.515
.550	-.103	-.159	-.151	-.155	-.424	-.157	-.243	-.219	-.206	-.465
.650	-.083	-.110	-.095	-.115	-.362	-.143	-.182	-.164	-.169	-.416
.750	-.053	-.053	-.034	-.075	-.304	-.082	-.080	-.085	-.127	-.359
.850	.013	.024	.030	-.019	-.219	.002	.024	.006	-.060	-.314
.900	.046	.068	.063	.016	-.171	.051	.078	.056	-.019	-.276
	<i>Right side</i>									
.025	-.376	.006	.323	.587	.823	-.949	-.042	.371	.599	.759
.075	-.389	-.100	.139	.354	.562	-.419	-.120	.169	.372	.550
.150	-.314	-.119	.065	.229	.399	-.328	-.135	.076	.240	.402
.250	-.226	-.115	.023	.158	.295	-.272	-.157	-.001	.141	.283
.350	-.205	-.143	-.034	.071	.189	-.267	-.196	-.069	.061	.190
.450	-.196	-.170	-.091	-.006	.091	-.269	-.234	-.134	-.029	.086
.550	-.155	-.163	-.108	-.040	.036	-.229	-.231	-.156	-.078	.019
.650	-.093	-.106	-.078	-.037	.008	-.158	-.176	-.131	-.087	-.020
.750	-.029	-.049	-.047	-.036	-.012	-.069	-.066	-.068	-.064	-.040
.850	.040	.035	.025	.017	-.012	.022	.041	.022	-.009	-.047
.900	.073	.063	.048	.025	-.026	.059	.068	.040	-.010	-.087
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.522	.793	.573	.176	-.178	.717	.865	.628	.251	-.255
.025	.384	-.089	-1.155	-.967	-.595	.438	-.254	-.810	-.859	-.421
.075	.207	-.130	-.416	-.902	-.601	.158	-.458	-.352	-.303	-.380
.150	.090	-.170	-.346	-.761	-.618	-.083	-.471	-.349	-.296	-.363
.250	-.002	-.226	-.289	-.534	-.612	-.155	-.358	-.348	-.300	-.372
.350	-.110	-.307	-.293	-.342	-.609					
.450	-.208	-.367	-.308	-.266	-.536					
.550	-.252	-.362	-.283	-.257	-.443					
.650	-.211	-.209	-.221	-.269	-.423	-.219	-.153	-.308	-.340	-.413
.750	-.123	-.090	-.164	-.281	-.424	-.147	-.041	-.302	-.350	-.407
.850	-.039	.019	-.104	-.279	-.411	-.083	.066	-.263	-.342	-.378
.900	-.009	.059	-.065	-.267	-.400	-.066	.100	-.228	-.328	-.361
	<i>Right side</i>									
.025	-1.114	-.029	.424	.639	.764	-.879	-.119	.488	.773	.953
.075	-.356	-.112	.207	.409	.584	-.409	-.427	.145	.436	.663
.150	-.322	-.135	.106	.287	.447	-.399	-.380	-.042	.190	.411
.250	-.304	-.211	-.002	.165	.320					
.350	-.322	-.286	-.109	.044	.194					
.450	-.320	-.350	-.201	-.065	.077					
.550	-.280	-.345	-.252	-.148	-.024					
.650	-.216	-.268	-.213	-.166	-.082	-.300	-.197	-.216	-.217	-.150
.750	-.153	-.076	-.125	-.145	-.116	-.283	-.072	-.158	-.212	-.190
.850	-.060	.044	-.021	-.082	-.102	-.238	.053	-.074	-.146	-.144
.900	-.022	.081	.005	-.091	-.139	-.201	.099	-.068	-.163	-.176



TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(i)  $\alpha = 0^\circ$ ;  $M = 0.90$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.804	.687	.615	.401	.075	.587	.802	.673	.329	-.119
.025	.283	-.054	-.478	-.953	-.819	.337	-.063	-.950	-1.281	-.542
.075	.152	-.090	-.381	-.751	-.688	.162	-.130	-.632	-1.166	-.540
.150	.045	-.147	-.368	-.660	-.637	.069	-.147	-.347	-1.055	-.550
.250	.016	-.119	-.238	-.469	-.558	-.008	-.166	-.272	-.484	-.556
.350	-.035	-.141	-.202	-.244	-.500	-.085	-.218	-.273	-.127	-.539
.450	-.105	-.190	-.203	-.164	-.477	-.176	-.289	-.281	-.208	-.507
.550	-.133	-.202	-.163	-.133	-.433	-.208	-.317	-.230	-.195	-.464
.650	-.125	-.161	-.103	-.111	-.381	-.219	-.279	-.170	-.167	-.422
.750	-.095	-.078	-.041	-.069	-.327	-.147	-.101	-.096	-.124	-.371
.850	-.019	.019	.030	-.013	-.237	-.031	.034	-.006	-.063	-.327
.900	.027	.070	.066	.019	-.217	.028	.090	.048	-.016	-.289
<i>Right side</i>										
.025	-.346	.019	.327	.595	.834	-.916	-.037	.368	.606	.768
.075	-.417	-.094	.142	.364	.571	-.614	-.114	.166	.380	.560
.150	-.351	-.121	.060	.238	.410	-.348	-.129	.073	.250	.410
.250	-.245	-.115	.021	.167	.305	-.265	-.155	-.002	.147	.292
.350	-.214	-.145	-.041	.074	.195	-.264	-.205	-.077	.060	.190
.450	-.203	-.185	-.107	-.008	.087	-.269	-.269	-.155	-.033	.084
.550	-.163	-.194	-.140	-.057	.026	-.233	-.290	-.201	-.096	.006
.650	-.099	-.142	-.123	-.063	-.018	-.160	-.242	-.209	-.128	-.045
.750	-.034	-.067	-.093	-.072	-.042	-.075	-.087	-.140	-.121	-.078
.850	.040	.035	-.007	-.016	-.046	.012	.055	-.025	-.057	-.087
.900	.070	.071	.024	.000	-.070	.046	.083	.013	-.037	-.125
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	.571	.815	.637	.288	-.081	.763	.897	.691	.336	-.127
.025	.390	-.065	-1.060	-1.347	-.585	.455	-.182	-.825	-.837	-.454
.075	.215	-.110	-.860	-1.194	-.590	.171	-.430	-.285	-.297	-.386
.150	.097	-.149	-.287	-.734	-.605	-.083	-.486	-.284	-.287	-.370
.250	.000	-.207	-.227	-.506	-.602	-.139	-.453	-.287	-.291	-.372
.350	-.118	-.323	-.270	-.369	-.609					
.450	-.238	-.426	-.303	-.287	-.543					
.550	-.350	-.473	-.273	-.248	-.446					
.650	-.334	-.441	-.225	-.244	-.423	-.581	-.246	-.291	-.319	-.407
.750	-.254	-.066	-.179	-.270	-.430	-.191	-.133	-.295	-.329	-.409
.850	-.067	.038	-.134	-.281	-.422	-.094	-.011	-.274	-.322	-.390
.900	-.031	.081	-.101	-.276	-.418	-.087	.045	-.253	-.315	-.375
<i>Right side</i>										
.025	-1.047	-.011	.425	.645	.782	-.754	-.066	.500	.789	.970
.075	-.799	-.095	.213	.429	.594	-.311	-.453	.151	.449	.673
.150	-.252	-.118	.115	.300	.460	-.307	-.411	-.040	.209	.415
.250	-.252	-.202	.001	.178	.330					
.350	-.293	-.310	-.111	.051	.198					
.450	-.300	-.379	-.227	-.069	.066					
.550	-.269	-.426	-.327	-.171	-.048					
.650	-.218	-.382	-.338	-.247	-.131	-.293	-.247	-.562	-.454	-.240
.750	-.171	-.034	-.317	-.276	-.191	-.293	-.158	-.256	-.362	-.293
.850	-.099	.063	-.054	-.135	-.148	-.264	-.043	-.104	-.148	-.182
.900	-.071	.101	-.018	-.113	-.174	-.235	.026	-.087	-.153	-.211

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(j)  $\alpha = 0^\circ$ ;  $M = 0.92$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.819	.688	.629	.431	.104	.612	.806	.689	.364	-.090
.025	.277	-.060	-.452	-.920	-.819	.334	-.073	-.879	-1.220	-.563
.075	.151	-.094	-.374	-.729	-.682	.156	-.141	-.686	-1.114	-.559
.150	.039	-.158	-.375	-.653	-.627	.067	-.151	-.394	-1.018	-.559
.250	.009	-.126	-.256	-.536	-.548	-.010	-.168	-.271	-.689	-.552
.350	-.039	-.146	-.202	-.308	-.494	-.091	-.224	-.266	-.138	-.534
.450	-.115	-.201	-.205	-.176	-.469	-.189	-.297	-.275	-.164	-.497
.550	-.151	-.227	-.165	-.127	-.426	-.224	-.341	-.236	-.180	-.454
.650	-.153	-.196	-.107	-.105	-.375	-.264	-.352	-.170	-.163	-.418
.750	-.135	-.102	-.039	-.075	-.319	-.224	-.137	-.095	-.131	-.363
.850	-.043	.014	.029	-.019	-.254	-.066	.034	-.002	-.073	-.321
.900	.011	.065	.066	.016	-.209	.007	.089	.051	-.030	-.275
	<i>Right side</i>									
.025	-.334	.023	.331	.603	.838	-.885	-.036	.368	.611	.772
.075	-.412	-.094	.147	.372	.579	-.637	-.121	.171	.385	.565
.150	-.370	-.126	.064	.242	.417	-.396	-.134	.077	.256	.416
.250	-.268	-.117	.023	.171	.313	-.270	-.155	.002	.151	.296
.350	-.228	-.153	-.041	.079	.205	-.269	-.205	-.070	.064	.200
.450	-.216	-.192	-.104	-.011	.093	-.273	-.277	-.153	-.034	.085
.550	-.172	-.215	-.148	-.061	.032	-.250	-.320	-.206	-.098	.007
.650	-.107	-.168	-.137	-.072	-.008	-.172	-.293	-.231	-.142	-.048
.750	-.042	-.085	-.119	-.093	-.042	-.084	-.120	-.189	-.154	-.082
.850	.033	.027	-.021	-.038	-.049	.007	.053	-.054	-.100	-.096
.900	.062	.067	.012	-.019	-.069	.041	.080	-.001	-.072	-.132
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.592	.821	.656	.323	-.056	.780	.912	.708	.368	-.090
.025	.389	-.071	-.984	-1.219	-.592	.464	-.168	-.796	-.779	-.455
.075	.217	-.114	-.830	-1.079	-.598	.172	-.427	-.271	-.298	-.377
.150	.096	-.151	-.428	-.850	-.612	-.086	-.497	-.271	-.289	-.361
.250	.002	-.205	-.172	-.517	-.603	-.133	-.472	-.272	-.289	-.363
.350	-.119	-.322	-.240	-.371	-.603					
.450	-.242	-.438	-.300	-.299	-.539					
.550	-.359	-.511	-.275	-.244	-.435					
.650	-.414	-.529	-.223	-.248	-.411	-.642	-.277	-.277	-.316	-.402
.750	-.380	-.094	-.178	-.273	-.419	-.520	-.199	-.284	-.327	-.409
.850	-.144	.040	-.136	-.290	-.417	-.135	-.081	-.266	-.318	-.387
.900	-.056	.080	-.104	-.289	-.413	-.103	-.023	-.246	-.311	-.375
	<i>Right side</i>									
.025	-1.003	-.006	.433	.655	.784	-.719	-.044	.508	.797	.975
.075	-.792	-.098	.221	.435	.599	-.301	-.445	.162	.455	.675
.150	-.301	-.117	.121	.307	.464	-.294	-.410	-.038	.213	.418
.250	-.222	-.198	.010	.180	.337					
.350	-.277	-.312	-.103	.053	.207					
.450	-.313	-.401	-.226	-.072	.069					
.550	-.275	-.458	-.328	-.175	-.046					
.650	-.220	-.426	-.378	-.254	-.133	-.264	-.254	-.617	-.495	-.281
.750	-.180	-.088	-.410	-.319	-.213	-.291	-.187	-.478	-.512	-.343
.850	-.112	.072	-.109	-.265	-.165	-.268	-.085	-.133	-.270	-.198
.900	-.084	.098	-.027	-.186	-.185	-.241	-.024	-.099	-.168	-.224

TABLE VIII - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(k)  $\alpha = 9^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.472	.822	.580	.036	-1.140	.018	.646	-.096	-1.036	-1.402
.025	-.085	-.102	-.048	-.446	-.925	.366	-.029	-.768	-1.294	-1.161
.075	.041	-.101	-.163	-.395	-.631	.210	-.050	-.402	-1.402	-1.127
.150	.055	-.115	-.200	-.342	-.480	.137	-.060	-.265	-.381	-1.108
.250	.055	-.085	-.138	-.234	-.331	.080	-.060	-.214	-.243	-.796
.350	.027	-.076	-.133	-.192	-.276	.038	-.078	-.194	-.243	-.480
.450	.004	-.090	-.133	-.167	-.246	.011	-.092	-.170	-.206	-.347
.550	-.001	-.078	-.110	-.130	-.216	.013	-.060	-.124	-.151	-.296
.650	.016	-.039	-.059	-.091	-.175	.018	-.039	-.082	-.107	-.267
.750	.029	-.016	-.025	-.036	-.136	.036	.003	-.041	-.059	-.228
.850	.057	.024	.029	.024	-.058	.077	.066	.040	.004	-.157
.900	.075	.047	.054	.066	-.007	.096	.091	.077	.043	-.111
<i>Right side</i>										
.025	.027	-.064	-.020	.211	.717	-.601	-.002	.422	.577	.660
.075	-.163	-.122	.047	.204	.541	-.356	-.036	.239	.405	.536
.150	-.186	-.104	.068	.211	.424	-.255	-.046	.156	.292	.417
.250	-.140	-.081	.068	.213	.343	-.195	-.050	.098	.211	.321
.350	-.138	-.076	.040	.170	.254	-.175	-.057	.063	.160	.254
.450	-.134	-.085	.015	.123	.183	-.159	-.062	.029	.103	.174
.550	-.101	-.074	.008	.100	.149	-.120	-.053	.019	.077	.132
.650	-.058	-.041	.017	.091	.128	-.072	-.027	.029	.068	.103
.750	-.021	-.006	.040	.087	.110	-.017	.024	.052	.073	.080
.850	.038	.038	.070	.112	.114	.052	.080	.093	.100	.077
.900	.061	.049	.089	.114	.112	.075	.084	.100	.091	.050
$z/b_v = 0.66$					$z/b_v = 0.93$					
<i>Left side</i>										
.000	-.040	.604	-.117	-.683	-.918	.337	.799	-.214	-.404	-.425
.025	.412	-.036	-.872	-.883	-.615	.662	.059	-.925	-.446	-.397
.075	.270	-.041	-.552	-.849	-.608	.449	-.034	-.849	-.436	-.411
.150	.183	-.041	-.316	-.805	-.590	.233	-.076	-.714	-.413	-.407
.250	.130	-.039	-.235	-.674	-.560	.137	-.048	-.471	-.381	-.402
.350	.084	-.062	-.210	-.457	-.540					
.450	.041	-.078	-.194	-.273	-.503					
.550	.006	-.081	-.157	-.185	-.455					
.650	.013	-.046	-.099	-.137	-.423	-.019	-.060	-.096	-.234	-.345
.750	.036	-.002	-.032	-.100	-.381	.009	-.013	-.032	-.206	-.331
.850	.055	.045	.024	-.068	-.349	.036	.033	.031	-.178	-.308
.900	.075	.073	.059	-.042	-.333	.057	.059	.061	-.162	-.299
<i>Right side</i>										
.025	-.792	.024	.457	.607	.655	-.962	.121	.718	.872	.963
.075	-.386	-.018	.290	.451	.568	-.716	.010	.478	.651	.802
.150	-.257	-.016	.212	.349	.474	-.689	-.025	.281	.437	.580
.250	-.218	-.027	.156	.280	.392					
.350	-.207	-.048	.093	.195	.291					
.450	-.186	-.067	.056	.133	.210					
.550	-.154	-.062	.017	.077	.130					
.650	-.092	-.036	.015	.047	.068	-.085	-.064	.003	.015	-.005
.750	-.028	-.009	.040	.045	.025	-.019	-.011	.003	-.031	-.099
.850	.048	.084	.091	.064	-.005	.059	.063	.066	-.022	-.136
.900	.077	.093	.098	.068	-.046	.087	.096	.087	-.038	-.182

TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(i)  $\alpha = 9^\circ$ ;  $M = 0.80$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.479	.890	.596	.235	-.496	.276	.692	.241	-.402	-.970
.025	-.107	-.108	.022	-.365	-.731	.379	-.032	-.837	-1.294	-.976
.075	.046	-.101	-.141	-.389	-.630	.224	-.058	-.424	-1.279	-.991
.150	.065	-.116	-.219	-.377	-.493	.148	-.054	-.267	-.647	-.947
.250	.064	-.086	-.152	-.253	-.377	.099	-.058	-.215	-.211	-.807
.350	.040	-.073	-.151	-.202	-.304	.050	-.081	-.207	-.236	-.568
.450	.011	-.095	-.155	-.181	-.268	.017	-.099	-.192	-.210	-.399
.550	.002	-.078	-.118	-.138	-.224	.017	-.072	-.143	-.156	-.318
.650	.012	-.041	-.063	-.087	-.190	.020	-.046	-.089	-.109	-.288
.750	.029	-.019	-.027	-.047	-.161	.040	.004	-.027	-.060	-.251
.850	.069	.032	.033	.017	-.096	.090	.074	.050	.003	-.193
.900	.096	.065	.065	.060	-.044	.117	.111	.093	.046	-.146
	<i>Right side</i>									
.025	.102	-.041	-.044	.220	.572	-.674	.027	.433	.585	.690
.075	-.132	-.116	.050	.212	.514	-.350	-.026	.247	.411	.559
.150	-.184	-.093	.079	.186	.443	-.242	-.032	.166	.305	.446
.250	-.137	-.069	.079	.197	.370	-.190	-.038	.115	.233	.353
.350	-.145	-.073	.048	.158	.280	-.180	-.054	.079	.170	.279
.450	-.148	-.092	.015	.117	.201	-.169	-.067	.042	.109	.201
.550	-.107	-.075	.008	.095	.158	-.131	-.060	.025	.077	.148
.650	-.058	-.046	.022	.083	.123	-.072	-.029	.030	.063	.110
.750	-.014	-.014	.033	.080	.110	-.001	.020	.059	.062	.079
.850	.050	.045	.080	.105	.107	.075	.096	.105	.092	.084
.900	.079	.058	.094	.106	.093	.108	.108	.105	.083	.027
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.219	.655	.167	-.303	-.811	.562	.878	.187	-.110	-.638
.025	.423	-.023	-.808	-.714	-.589	.677	.020	-.849	-.432	-.393
.075	.288	-.032	-.592	-.685	-.575	.470	-.063	-.741	-.395	-.402
.150	.213	-.032	-.339	-.632	-.556	.247	-.115	-.710	-.357	-.400
.250	.169	-.031	-.241	-.575	-.522	.140	-.078	-.600	-.342	-.394
.350	.104	-.064	-.238	-.512	-.504					
.450	.041	-.096	-.229	-.382	-.486					
.550	.002	-.101	-.186	-.260	-.449					
.650	.005	-.063	-.109	-.181	-.425	-.037	-.069	-.076	-.224	-.361
.750	.032	-.002	-.027	-.139	-.399	-.005	-.003	-.007	-.205	-.349
.850	.065	.062	.042	-.110	-.373	.040	.067	.054	-.191	-.332
.900	.090	.088	.079	-.096	-.362	.069	.100	.086	-.179	-.321
	<i>Right side</i>									
.025	-.799	.064	.475	.626	.706	-.892	.149	.742	.904	1.030
.075	-.425	.007	.310	.477	.615	-.644	.004	.496	.677	.851
.150	-.231	.009	.241	.379	.519	-.661	-.022	.296	.451	.624
.250	-.215	-.014	.178	.309	.441					
.350	-.222	-.044	.115	.217	.332					
.450	-.204	-.075	.053	.137	.239					
.550	-.158	-.081	.010	.074	.148					
.650	-.085	-.054	.015	.036	.075	.015	-.031	-.010	-.020	-.008
.750	-.011	-.017	.034	.017	.012	.017	.003	.001	-.081	-.125
.850	.082	.105	.089	.032	-.023	.087	.087	.062	-.081	-.184
.900	.117	.117	.106	.029	-.079	.120	.119	.083	-.104	-.253

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(m)  $\alpha = 9^\circ$ ;  $M = 0.85$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.478	.901	.287	.274	-.384	.327	.707	-.233	-.260	-.947
.025	-.124	-.109	-.321	-.334	-.548	.370	-.041	-1.300	-1.308	-.938
.075	.036	-.105	-.377	-.388	-.611	.219	-.068	-1.183	-1.152	-.921
.150	.060	-.121	-.407	-.422	-.517	.149	-.064	-.719	-.766	-.912
.250	.063	-.094	-.271	-.281	-.382	.097	-.067	-.234	-.267	-.811
.350	.032	-.084	-.214	-.225	-.328	.048	-.089	-.264	-.277	-.612
.450	.000	-.104	-.200	-.213	-.291	.009	-.109	-.246	-.257	-.431
.550	-.008	-.087	-.164	-.173	-.238	.006	-.087	-.199	-.213	-.336
.650	-.001	-.048	-.119	-.131	-.204	.009	-.057	-.159	-.175	-.296
.750	.018	-.029	-.079	-.091	-.168	.029	-.008	-.109	-.122	-.259
.850	.060	.032	-.004	-.017	-.108	.080	.073	-.032	-.044	-.202
.900	.089	.063	.042	.031	-.053	.109	.112	.019	-.004	-.151
	<i>Right side</i>									
.025	.122	-.042	.212	.209	.547	-.806	.035	.588	.594	.697
.075	-.136	-.118	.210	.212	.473	-.365	-.027	.410	.415	.584
.150	-.202	-.099	.183	.185	.433	-.253	-.035	.310	.305	.451
.250	-.154	-.072	.190	.193	.378	-.204	-.044	.236	.236	.358
.350	-.161	-.077	.157	.156	.290	-.192	-.058	.173	.170	.291
.450	-.168	-.101	.109	.105	.209	-.191	-.077	.110	.105	.212
.550	-.128	-.082	.088	.084	.165	-.151	-.072	.075	.068	.158
.650	-.071	-.051	.073	.068	.126	-.091	-.038	.056	.050	.115
.750	-.025	-.021	.076	.065	.108	-.011	.009	.052	.047	.085
.850	.042	.040	.096	.088	.103	.069	.093	.072	.064	.062
.900	.072	.063	.095	.088	.089	.103	.106	.065	.051	.025
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.270	.672	-.184	-.212	-.749	.624	.888	-.062	-.164	-.615
.025	.426	-.028	-.636	-.645	-.679	.682	-.015	-.499	-.523	-.413
.075	.287	-.035	-.616	-.627	-.625	.468	-.081	-.417	-.415	-.416
.150	.212	-.035	-.563	-.582	-.588	.239	-.139	-.367	-.371	-.411
.250	.166	-.038	-.482	-.494	-.554	.136	-.101	-.358	-.367	-.410
.350	.096	-.081	-.434	-.436	-.524					
.450	.032	-.116	-.401	-.399	-.497					
.550	-.018	-.124	-.355	-.361	-.460					
.650	-.019	-.081	-.304	-.315	-.430	-.065	-.084	-.327	-.342	-.374
.750	.012	-.015	-.249	-.271	-.410	-.029	-.014	-.297	-.323	-.362
.850	.052	.055	-.186	-.213	-.384	.022	.070	-.251	-.283	-.347
.900	.079	.093	-.153	-.198	-.373	.050	.103	-.226	-.260	-.336
	<i>Right side</i>									
.025	-.850	.072	.632	.638	.722	-.855	.159	.915	.919	1.045
.075	-.453	.018	.485	.486	.627	-.640	-.007	.686	.686	.866
.150	-.236	.016	.387	.389	.532	-.631	-.027	.459	.457	.640
.250	-.221	-.018	.316	.312	.455					
.350	-.245	-.052	.222	.221	.351					
.450	-.238	-.089	.140	.128	.251					
.550	-.185	-.098	.066	.062	.160					
.650	-.105	-.068	.024	.018	.086	-.055	-.048	-.039	-.044	-.013
.750	-.022	-.022	-.008	-.019	.020	.000	.003	-.116	-.129	-.111
.850	.070	.103	.008	-.007	-.026	.068	.095	-.116	-.135	-.165
.900	.107	.120	-.002	-.024	-.080	.099	.129	-.146	-.178	-.244

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(n)  $\alpha = 9^\circ$ ;  $M = 0.90$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.496	.926	.618	.340	-.417	.369	.712	.408	-.078	-.723
.025	-.115	-.117	.080	-.262	-.518	.380	-.053	-.978	-1.355	-.980
.075	.044	-.116	-.111	-.333	-.554	.228	-.080	-.426	-1.042	-.935
.150	.070	-.135	-.238	-.424	-.585	.157	-.077	-.268	-.489	-.888
.250	.072	-.107	-.160	-.285	-.382	.109	-.078	-.207	-.332	-.795
.350	.041	-.096	-.152	-.214	-.348	.056	-.105	-.204	-.272	-.632
.450	.012	-.125	-.166	-.201	-.329	.019	-.140	-.198	-.242	-.480
.550	.005	-.111	-.132	-.165	-.274	.012	-.120	-.158	-.205	-.384
.650	.012	-.072	-.081	-.123	-.235	.013	-.092	-.111	-.170	-.337
.750	.021	-.052	-.043	-.084	-.201	.034	-.036	-.048	-.119	-.293
.850	.066	.013	-.019	-.013	-.130	.087	.054	.036	-.047	-.235
.900	.095	.046	.060	.032	-.091	.115	.099	.084	.005	-.188
	<i>Right side</i>									
.025	.148	-.048	-.059	.203	.484	-.839	.028	.431	.592	.710
.075	-.118	-.134	.042	.210	.486	-.348	-.037	.250	.414	.568
.150	-.188	-.111	.079	.189	.455	-.244	-.044	.171	.312	.458
.250	-.140	-.089	.082	.186	.391	-.188	-.054	.122	.241	.366
.350	-.146	-.089	.046	.155	.296	-.179	-.074	.080	.182	.299
.450	-.156	-.113	.007	.108	.210	-.178	-.099	.031	.114	.218
.550	-.112	-.097	.002	.088	.165	-.143	-.096	.008	.071	.159
.650	-.059	-.074	-.001	.072	.119	-.082	-.070	.007	.051	.119
.750	-.016	-.042	.020	.064	.096	.002	-.015	.030	.045	.080
.850	.058	.028	.063	.087	.089	.081	.081	.076	.060	.053
.900	.086	.048	.079	.084	.067	.109	.093	.084	.045	.010
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.307	.680	.318	-.123	-.677	.664	.913	.457	-.025	-.610
.025	.437	-.033	-.982	-.706	-.720	.687	-.019	-.711	-.758	-.452
.075	.305	-.038	-.563	-.657	-.653	.476	-.107	-.397	-.357	-.460
.150	.226	-.042	-.260	-.594	-.612	.243	-.174	-.358	-.328	-.449
.250	.179	-.056	-.187	-.482	-.586	.140	-.128	-.350	-.324	-.451
.350	.108	-.111	-.198	-.392	-.546					
.450	.035	-.158	-.218	-.327	-.526					
.550	-.015	-.174	-.210	-.272	-.495					
.650	-.018	-.131	-.153	-.249	-.471	-.065	-.123	-.240	-.352	-.417
.750	.010	-.045	-.087	-.244	-.448	-.030	-.029	-.196	-.353	-.404
.850	.056	.040	-.014	-.216	-.424	.017	.058	-.137	-.327	-.386
.900	.081	.076	.028	-.195	-.416	.042	.093	-.101	-.308	-.374
	<i>Right side</i>									
.025	-.840	.069	.483	.643	.736	-.794	.134	.732	.930	1.061
.075	-.437	.010	.321	.494	.638	-.575	-.040	.484	.697	.882
.150	-.217	.009	.253	.402	.540	-.560	-.050	.285	.470	.651
.250	-.204	-.026	.190	.329	.466					
.350	-.232	-.078	.118	.232	.360					
.450	-.234	-.124	.042	.142	.262					
.550	-.186	-.147	-.020	.082	.167					
.650	-.104	-.121	-.040	.011	.089	-.066	-.076	-.071	-.059	.010
.750	-.016	-.062	-.028	-.027	.016	-.009	-.025	-.095	-.134	-.097
.850	.083	.089	.036	-.013	-.022	.059	.084	-.047	-.134	-.132
.900	.118	.108	.056	-.032	-.078	.091	.115	-.041	-.186	-.205

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(o)  $\alpha = 9^\circ$ ;  $M = 0.92$

$x$ $C_v$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.491	.935	.621	.346	-.383	.415	.716	.435	-.022	-.672
.025	-.143	-.101	.086	-.248	-.481	.374	-.037	-.990	-1.340	-1.013
.075	.028	-.102	-.106	-.325	-.531	.220	-.069	-.428	-.964	-.959
.150	.059	-.125	-.247	-.437	-.583	.148	-.065	-.277	-.586	-.887
.250	.057	-.095	-.169	-.319	-.405	.100	-.069	-.211	-.374	-.789
.350	.032	-.084	-.162	-.239	-.342	.045	-.095	-.211	-.329	-.629
.450	-.006	-.117	-.176	-.231	-.336	-.001	-.134	-.204	-.271	-.487
.550	-.021	-.108	-.143	-.191	-.278	-.013	-.122	-.172	-.229	-.397
.650	-.016	-.071	-.091	-.148	-.253	-.022	-.096	-.124	-.194	-.358
.750	-.004	-.046	-.052	-.111	-.219	-.006	-.039	-.061	-.145	-.313
.850	.044	.020	.008	-.037	-.154	.053	.060	.028	-.069	-.253
.900	.078	.059	.047	.009	-.105	.090	.106	.074	-.021	-.204
	<i>Right side</i>									
.025	.163	-.031	-.077	.183	.464	-.872	.035	.420	.581	.713
.075	-.112	-.122	.029	.191	.481	-.355	-.027	.242	.400	.571
.150	-.212	-.096	.070	.176	.458	-.259	-.033	.161	.296	.458
.250	-.159	-.074	.071	.167	.396	-.195	-.040	.111	.228	.367
.350	-.166	-.074	.037	.138	.298	-.185	-.062	.073	.171	.298
.450	-.173	-.108	.002	.095	.213	-.189	-.091	.024	.101	.219
.550	-.133	-.089	-.007	.070	.168	-.164	-.102	-.003	.062	.163
.650	-.081	-.070	-.013	.055	.118	-.105	-.072	-.006	.037	.117
.750	-.038	-.041	.008	.047	.097	-.026	-.018	.015	.025	.080
.850	.036	.032	.052	.068	.083	.057	.082	.064	.039	.050
.900	.062	.056	.070	.062	.061	.092	.102	.073	.024	.005
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	.353	.692	.347	-.088	-.649	.723	.922	.489	.007	-.593
.025	.430	-.017	-1.015	-.804	-.734	.684	.011	-.701	-.845	-.449
.075	.298	-.027	-.546	-.721	-.658	.463	-.105	-.371	-.383	-.471
.150	.223	-.030	-.261	-.643	-.616	.228	-.173	-.344	-.354	-.462
.250	.162	-.041	-.187	-.528	-.592	.123	-.110	-.338	-.351	-.445
.350	.092	-.102	-.198	-.401	-.556					
.450	.012	-.157	-.229	-.322	-.531					
.550	-.052	-.179	-.220	-.274	-.506					
.650	-.068	-.143	-.166	-.261	-.481	-.135	-.128	-.245	-.377	-.430
.750	-.042	-.041	-.102	-.258	-.465	-.105	-.027	-.214	-.386	-.417
.850	.007	.045	-.030	-.236	-.440	-.075	.067	-.165	-.365	-.401
.900	.033	.086	.010	-.221	-.433	-.062	.106	-.133	-.347	-.395
	<i>Right side</i>									
.025	-.956	.084	.480	.639	.742	-.650	.173	.727	.930	1.070
.075	-.397	.025	.313	.482	.641	-.380	-.032	.476	.690	.887
.150	-.192	.025	.252	.394	.547	-.363	-.046	.278	.457	.658
.250	-.172	-.014	.182	.316	.469					
.350	-.208	-.061	.106	.223	.364					
.450	-.229	-.118	.031	.126	.265					
.550	-.209	-.147	-.035	.043	.166					
.650	-.141	-.128	-.060	-.016	.090	-.197	-.078	-.097	-.079	.016
.750	-.068	-.071	-.051	-.049	.017	-.159	-.010	-.118	-.153	-.098
.850	.027	.097	.020	-.030	-.020	-.112	.089	-.069	-.145	-.123
.900	.065	.116	.039	-.057	-.075	-.073	.119	-.073	-.203	-.193

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

(p)  $\alpha = 15.1^\circ$ ;  $M = 0.60$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.408	.851	.706	.595	.309	-.314	.796	-.232	-1.556	-2.058
.025	-.450	-.207	.174	-.030	-.449	.418	.038	-.643	-2.050	-2.060
.075	-.151	-.177	-.043	-.155	-.449	.275	-.009	-.373	-.526	-2.364
.150	-.035	-.167	-.195	-.241	-.447	.208	-.001	-.251	-.424	-.490
.250	.033	-.131	-.179	-.180	-.312	.156	-.003	-.200	-.348	-.529
.350	.021	-.089	-.177	-.157	-.252	.112	-.024	-.191	-.317	-.472
.450	.003	-.107	-.186	-.159	-.215	.082	-.045	-.170	-.285	-.421
.550	-.016	-.098	-.158	-.125	-.167	.077	-.024	-.133	-.250	-.376
.650	-.035	-.068	-.112	-.084	-.110	.070	-.008	-.094	-.230	-.337
.750	-.025	-.040	-.075	-.057	-.067	.077	.022	-.040	-.187	-.284
.850	-.002	-.003	-.027	-.016	-.007	.098	.066	.015	-.134	-.238
.900	.017	.017	.008	.014	.032	.110	.089	.057	-.084	-.193
	<i>Right side</i>									
.025	.264	-.121	-.371	.016	.208	-.778	.024	.412	.517	.574
.075	-.053	-.191	-.172	.055	.235	-.382	-.017	.267	.401	.524
.150	-.186	-.161	-.040	.096	.251	-.258	-.013	.195	.319	.441
.250	-.179	-.117	.027	.128	.201	-.203	-.013	.154	.260	.366
.350	-.182	-.089	.027	.105	.222	-.177	-.029	.117	.210	.304
.450	-.186	-.112	.001	.068	.217	-.165	-.031	.089	.164	.245
.550	-.154	-.091	.001	.048	.185	-.135	-.031	.068	.132	.194
.650	-.105	-.066	.001	.043	.158	-.095	-.010	.068	.114	.155
.750	-.060	-.031	-.001	.057	.139	-.037	.024	.087	.109	.128
.850	-.007	.013	.017	.078	.130	.028	.073	.112	.119	.110
.900	.012	.015	.020	.078	.119	.054	.084	.114	.116	.087
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	-.335	.578	-.345	-1.082	-1.005	.180	.863	-.359	-.622	-.554
.025	.473	.068	-.784	-1.016	-.980	.821	.193	-.717	-.435	-.598
.075	.357	.059	-.590	-1.034	-.984	.627	.158	-.687	-.414	-.536
.150	.280	.061	-.255	-1.052	-.952	.413	.112	-.551	-.421	-.529
.250	.240	.059	-.140	-.629	-.893	.285	.089	-.401	-.405	-.463
.350	.189	.045	-.131	-.102	-.721					
.450	.131	.015	-.126	-.084	-.540					
.550	.084	.003	-.112	-.118	-.419					
.650	.068	.020	-.061	-.100	-.334	.021	-.008	-.040	-.184	-.302
.750	.068	.050	-.008	-.071	-.284	.000	.006	-.008	-.162	-.286
.850	.068	.073	.038	-.032	-.231	.012	.036	.031	-.134	-.296
.900	.075	.089	.066	.000	-.202	.024	.063	.052	-.114	-.300
	<i>Right side</i>									
.025	-.778	.075	.488	.581	.613	-.764	.239	.849	.961	1.009
.075	-.596	.052	.357	.492	.602	-.725	.154	.606	.770	.897
.150	-.247	.066	.292	.419	.535	-.485	.135	.433	.570	.704
.250	-.130	.061	.237	.349	.455					
.350	-.128	.045	.181	.273	.373					
.450	-.114	.017	.130	.207	.286					
.550	-.093	.015	.096	.150	.206					
.650	-.053	.022	.075	.105	.135	-.023	.006	.038	.055	.041
.750	-.011	.043	.068	.075	.071	-.011	.006	.006	-.045	-.094
.850	.061	.093	.098	.078	.029	.047	.063	.036	-.066	-.174
.900	.084	.117	.112	.071	-.005	.070	.084	.047	-.091	-.257



TABLE VIII.- PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Continued

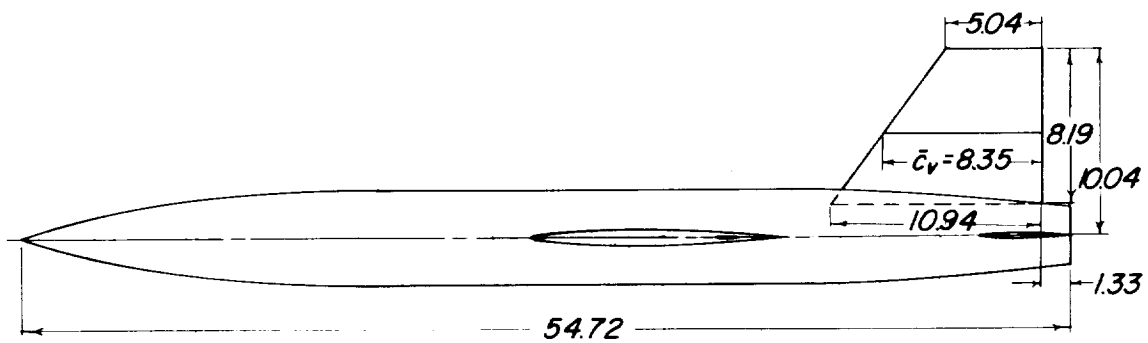
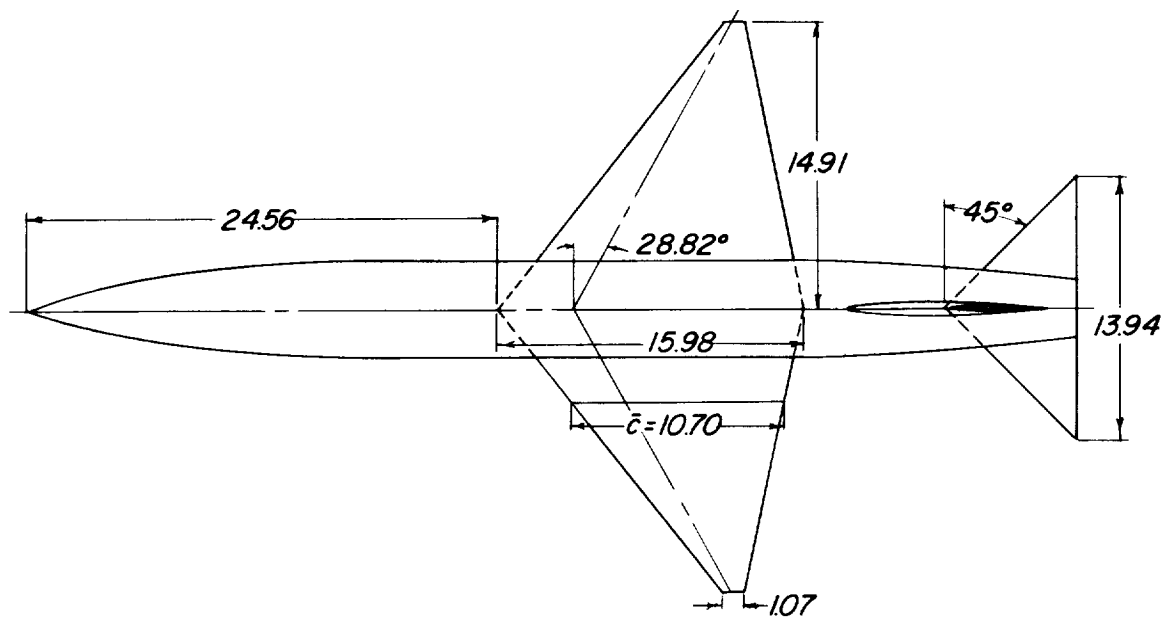
(q)  $\alpha = 15.1^\circ$ ;  $M = 0.80$

$\frac{x}{c_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
	<i>Left side</i>									
.000	.574	.905	.816	.714	.412	.110	.891	.206	-.774	-1.499
.025	-.592	-.191	.216	.079	-.248	.417	.056	-.564	-1.505	-1.703
.075	-.159	-.175	-.005	-.100	-.337	.283	.001	-.362	-.677	-1.819
.150	-.035	-.164	-.202	-.274	-.475	.227	.003	-.254	-.490	-.580
.250	.038	-.113	-.188	-.217	-.355	.174	.001	-.204	-.391	-.547
.350	.032	-.071	-.182	-.186	-.274	.129	-.016	-.191	-.343	-.567
.450	.006	-.098	-.202	-.192	-.298	.096	-.033	-.182	-.316	-.482
.550	-.009	-.086	-.167	-.150	-.203	.088	-.025	-.146	-.285	-.421
.650	-.028	-.059	-.112	-.101	-.148	.079	-.008	-.104	-.265	-.375
.750	-.028	-.036	-.069	-.072	-.103	.085	.024	-.043	-.223	-.319
.850	-.003	.010	-.019	-.027	-.031	.105	.073	.019	-.157	-.258
.900	.018	.033	.016	-.001	.016	.122	.097	.061	-.112	-.209
	<i>Right side</i>									
.025	.320	-.075	-.390	-.018	.178	-.703	.015	.403	.525	.603
.075	-.014	-.191	-.173	.032	.210	-.404	-.023	.267	.402	.541
.150	-.201	-.150	-.025	.084	.245	-.278	-.013	.213	.328	.463
.250	-.197	-.095	.044	.129	.222	-.215	-.005	.177	.272	.397
.350	-.198	-.071	.039	.112	.177	-.195	-.020	.140	.229	.305
.450	-.215	-.100	.021	.079	.189	-.182	-.037	.103	.186	.279
.550	-.171	-.078	.016	.061	.180	-.151	-.036	.088	.150	.232
.650	-.113	-.052	.010	.049	.161	-.107	-.016	.088	.132	.193
.750	-.067	-.026	.015	.059	.145	-.043	.022	.094	.120	.157
.850	-.003	.022	.029	.081	.138	.026	.076	.119	.121	.128
.900	.017	.041	.030	.072	.120	.058	.083	.122	.110	.091
	$z/b_v = 0.66$					$z/b_v = 0.93$				
	<i>Left side</i>									
.000	-.064	.624	-.065	-.830	-.964	.381	.922	-.034	-.573	-.640
.025	.494	.094	-.846	-1.038	-.933	.858	.192	-.765	-.573	-.669
.075	.376	.079	-.703	-1.053	-.952	.642	.146	-.684	-.524	-.599
.150	.306	.079	-.268	-1.027	-.909	.419	.097	-.610	-.527	-.588
.250	.270	.077	-.139	-.763	-.891	.297	.088	-.489	-.451	-.537
.350	.212	.054	-.150	-.197	-.759					
.450	.148	.018	-.158	-.117	-.599					
.550	.091	-.004	-.133	-.157	-.493					
.650	.067	.006	-.078	-.150	-.426	.003	-.031	-.059	-.274	-.381
.750	.059	.033	-.022	-.123	-.371	-.029	-.016	-.022	-.259	-.366
.850	.058	.065	.033	-.078	-.304	-.041	.024	.021	-.243	-.363
.900	.067	.085	.061	-.055	-.271	-.034	.050	.042	-.232	-.368
	<i>Right side</i>									
.025	-.810	.108	.520	.612	.652	-.872	.259	.882	1.019	1.086
.075	-.689	.082	.384	.516	.632	-.691	.158	.636	.814	.966
.150	-.305	.096	.323	.454	.577	-.593	.137	.448	.603	.762
.250	-.137	.088	.277	.388	.513					
.350	-.148	.054	.210	.309	.430					
.450	-.142	.030	.154	.235	.345					
.550	-.124	.007	.108	.167	.256					
.650	-.078	.013	.077	.112	.183	-.072	-.017	.021	.027	.085
.750	-.023	.038	.067	.066	.103	-.038	-.008	-.020	-.066	-.042
.850	.053	.096	.094	.056	.048	.023	.061	.004	-.107	-.122
.900	.081	.114	.099	.039	.001	.045	.079	.000	-.188	-.224

TABLE VIII. - PRESSURE COEFFICIENTS FOR THE COMBINATION OF THE FUSELAGE, VERTICAL TAIL, AND HORIZONTAL TAIL. HORIZONTAL TAIL IN THE FORWARD POSITION;  $i_t = 0^\circ$  - Concluded

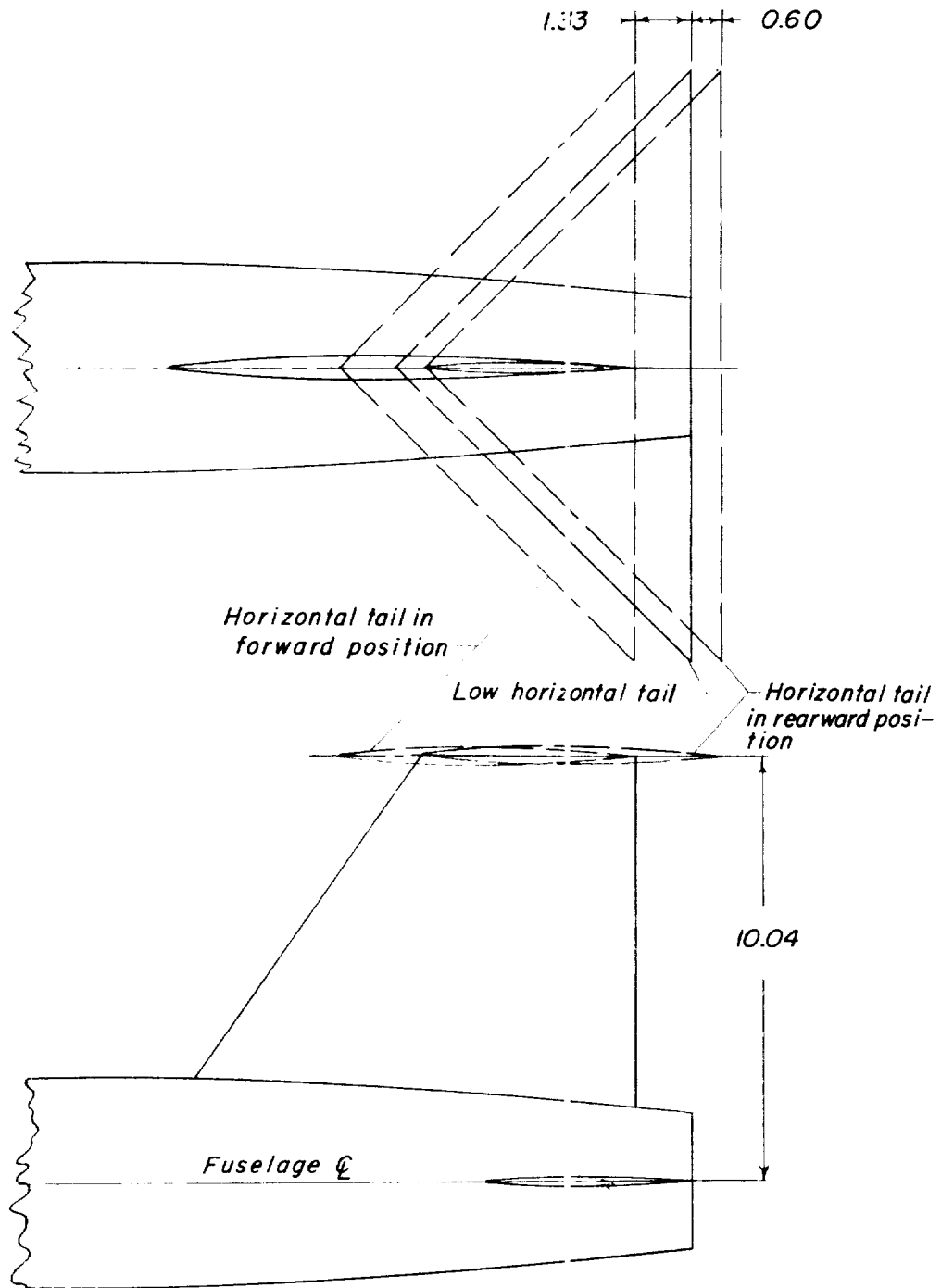
(r)  $\alpha = 15.1^\circ$ ;  $M = 0.85$

$\frac{x}{C_v}$	$C_p$ for -					$C_p$ for -				
	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$	$\beta = -3.9^\circ$	$\beta = 0^\circ$	$\beta = 3.9^\circ$	$\beta = 7.9^\circ$	$\beta = 12.7^\circ$
	$z/b_v = 0.11$					$z/b_v = 0.38$				
<i>Left side</i>										
.000	.637	.928	.851	.756	.417	.197	.888	.295	-.570	-1.255
.025	-.650	-.200	.225	.123	-.174	.410	.043	-.538	-1.351	-1.577
.075	-.146	-.195	.008	-.065	-.272	.284	-.019	-.376	-.706	-1.623
.150	-.035	-.179	-.208	-.280	-.450	.227	-.014	-.271	-.541	-.741
.250	.033	-.129	-.199	-.249	-.402	.177	-.013	-.215	-.427	-.422
.350	.030	-.087	-.194	-.220	-.296	.132	-.032	-.205	-.390	-.644
.450	.005	-.110	-.204	-.211	-.236	.095	-.054	-.197	-.358	-.503
.550	-.009	-.102	-.179	-.180	-.232	.088	-.046	-.165	-.327	-.467
.650	-.027	-.082	-.126	-.127	-.176	.080	-.027	-.121	-.304	-.417
.750	-.035	-.053	-.083	-.098	-.141	.080	.001	-.059	-.251	-.355
.850	-.007	-.003	-.028	-.043	-.059	.102	.054	.010	-.179	-.285
.900	.013	.020	.010	-.012	-.008	.115	.079	.054	-.122	-.231
<i>Right side</i>										
.025	.343	-.083	-.430	-.043	.165	-.694	.000	.394	.518	.611
.075	.005	-.216	-.178	.021	.190	-.413	-.044	.265	.405	.541
.150	-.209	-.163	-.028	.082	.232	-.293	-.027	.213	.331	.467
.250	-.211	-.110	.046	.128	.227	-.229	-.022	.179	.283	.404
.350	-.209	-.084	.041	.113	.170	-.204	-.039	.139	.207	.280
.450	-.226	-.112	.018	.082	.167	-.198	-.060	.110	.197	.290
.550	-.185	-.094	.014	.062	.169	-.168	-.056	.090	.162	.243
.650	-.119	-.070	.010	.053	.157	-.118	-.032	.087	.143	.202
.750	-.075	-.040	.014	.058	.142	-.054	.004	.096	.128	.185
.850	-.012	.010	.025	.078	.133	.020	.064	.116	.125	.132
.900	.013	.029	.033	.075	.117	.053	.066	.121	.109	.092
$z/b_v = 0.66$										
<i>Left side</i>										
.000	-.012	.634	.005	-.716	-.901	.440	.936	.081	-.606	-.716
.025	.503	.086	-.866	-1.105	-.884	.869	.194	-.750	-.714	-.701
.075	.384	.074	-.748	-1.113	-.912	.654	.124	-.707	-.672	-.646
.150	.320	.074	-.298	-1.033	-.897	.431	.083	-.632	-.682	-.644
.250	.279	.067	-.146	-.736	-.890	.306	.071	-.526	-.626	-.606
.350	.219	.041	-.165	-.377	-.777					
.450	.152	.001	-.179	-.220	-.647					
.550	.090	-.023	-.158	-.211	-.552					
.650	.066	-.017	-.098	-.192	-.487	-.002	-.064	-.081	-.321	-.450
.750	.052	.010	-.039	-.150	-.426	-.041	-.047	-.050	-.291	-.427
.850	.048	.043	.020	-.095	-.352	-.064	-.006	-.002	-.271	-.415
.900	.059	.060	.051	-.068	-.318	-.068	.023	.021	-.266	-.412
$z/b_v = 0.93$										
<i>Right side</i>										
.025	-.873	.100	.530	.620	.667	-.885	.255	.894	1.041	1.109
.075	-.791	.076	.388	.525	.640	-.699	.196	.643	.834	.978
.150	-.320	.089	.332	.464	.587	-.621	.123	.452	.623	.779
.250	-.145	.080	.288	.400	.523					
.350	-.165	.047	.220	.323	.441					
.450	-.166	.009	.162	.251	.360					
.550	-.138	-.014	.110	.179	.272					
.650	-.095	-.010	.077	.125	.199	-.091	-.050	.017	.052	.103
.750	-.037	.014	.063	.078	.119	-.061	-.040	-.033	-.049	-.017
.850	.046	.077	.091	.062	.059	-.001	.031	-.016	-.086	-.087
.900	.073	.094	.091	.041	-.001	.025	.049	-.029	-.169	-.185



(a) Complete model with horizontal tail in low position.

Figure 1.- Details of test model. (All linear dimensions are in inches.)

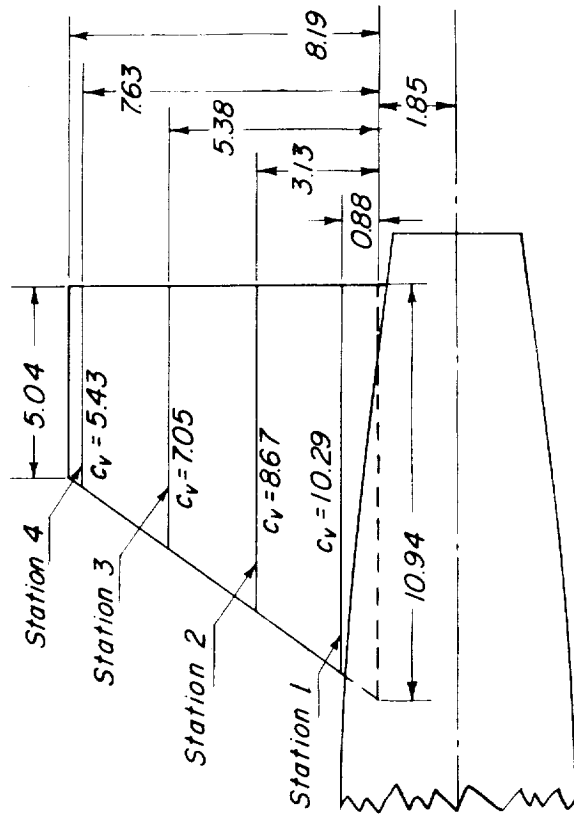


(b) Horizontal-tail positions.

Figure 1.- Continued.

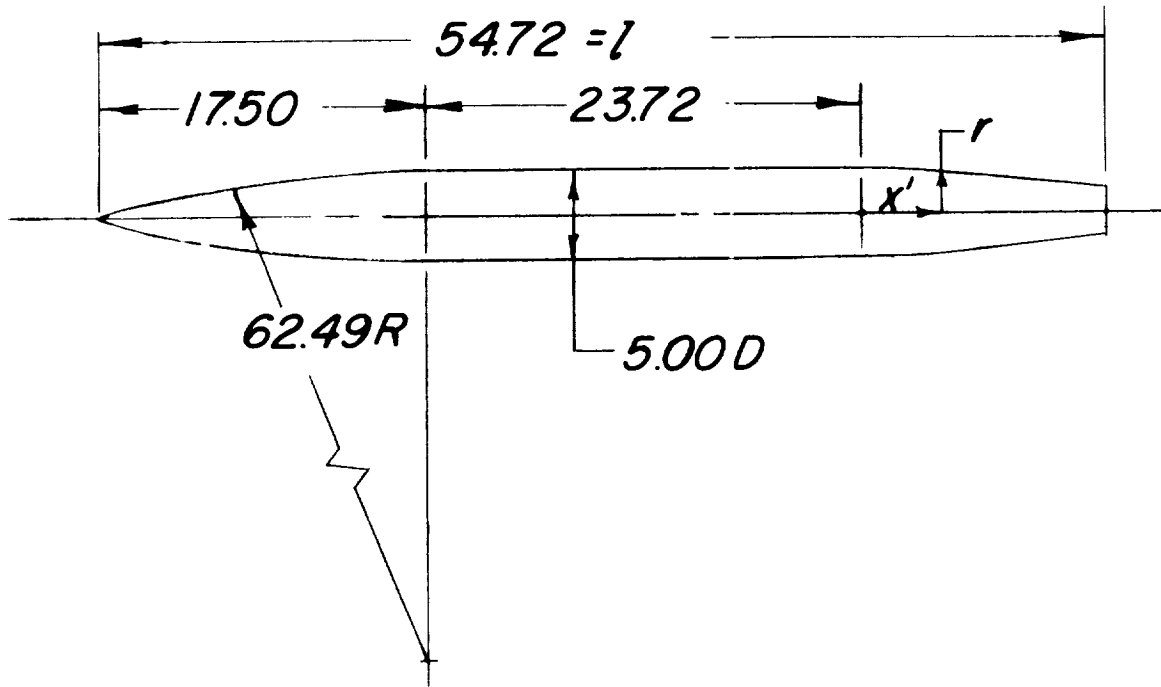
Station	$\frac{z}{b_v}$
1	0.11
2	0.38
3	0.66
4	0.93

Chordwise location of static orifices, $x_{c_v}$	Station 4	
	Left side	Right side
Stations 1, 2, 3		
Left and right sides		
.000	.000	.000
.025	.025	.025
.075	.075	.075
.150	.150	.150
.250	.250	.150
.350	—	—
.450	—	—
.550	—	—
.650	.650	.650
.750	.750	.750
.850	.850	.850
.900	.900	.900



(c) Vertical-tail pressure-orifice locations.

Figure 1.- Continued.



### Afterbody Coordinates

$x'/l$	$r'/r$
0	.0456
.0320	.0445
.0639	.0427
.1187	.0390
<i>Straight-line taper</i>	
.2460	.0301

(d) Fuselage dimensions; fineness ratio, 10.94.

Figure 1.- Concluded.

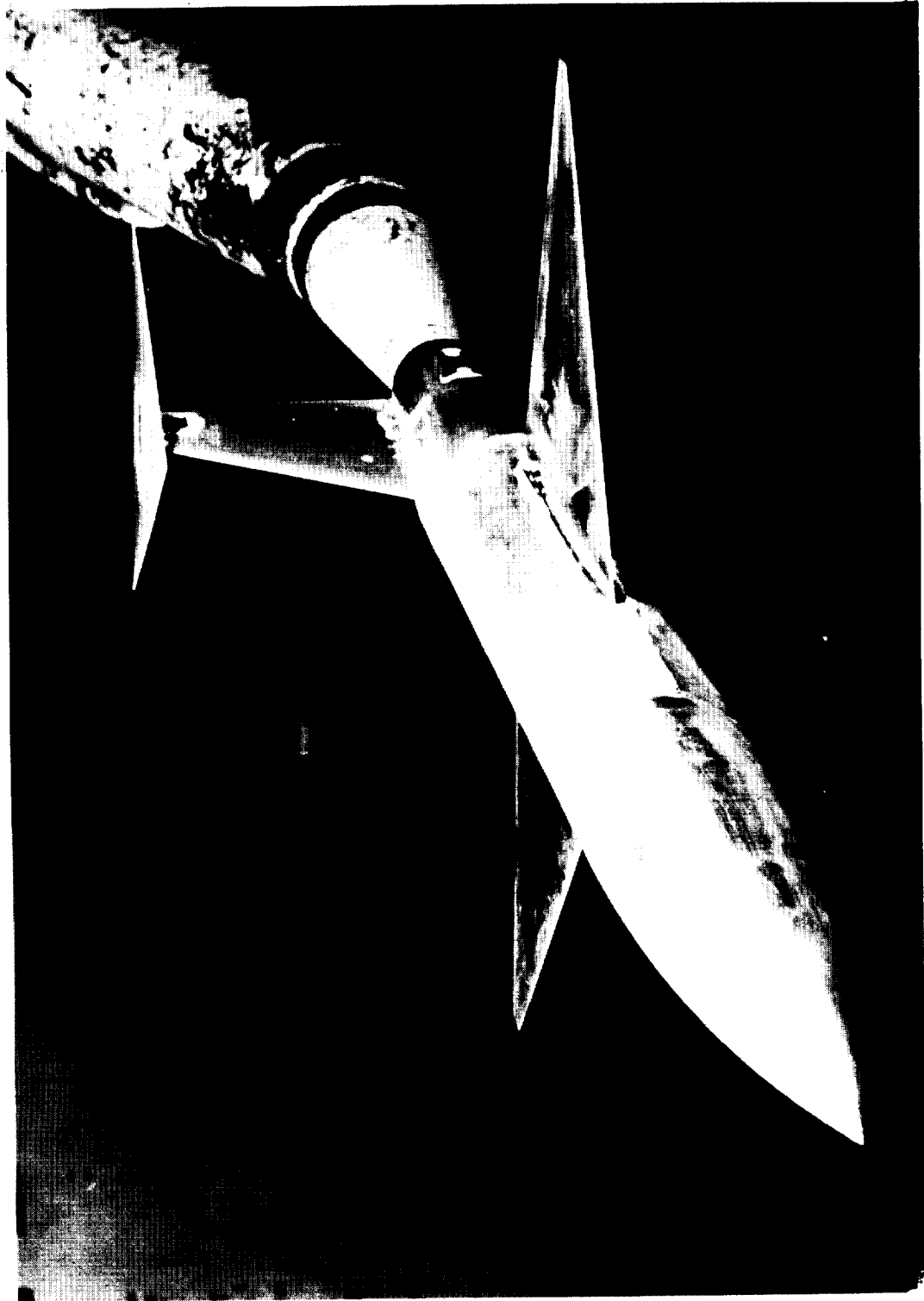


Figure 2.- Model mounted in tunnel. L-89250

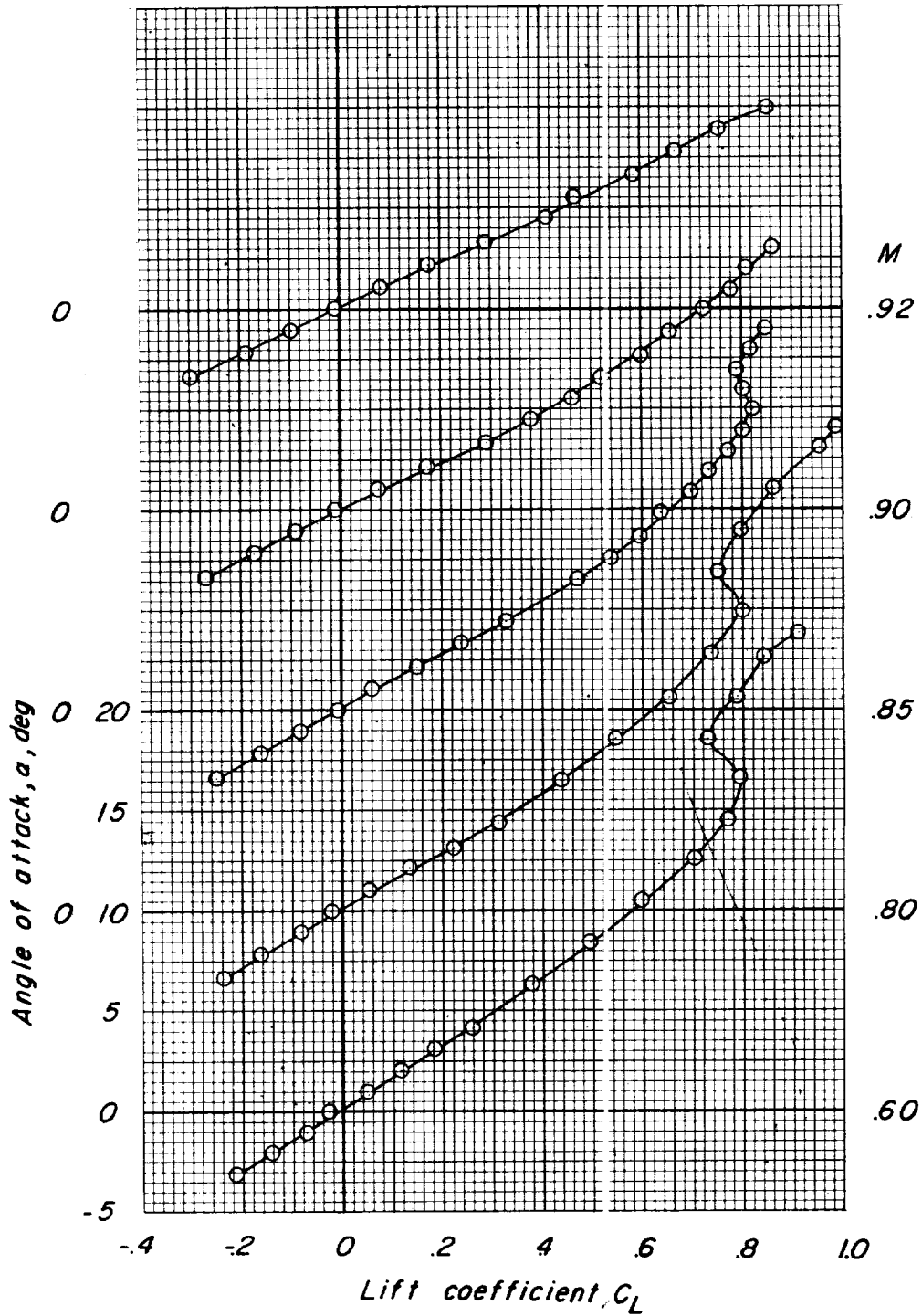
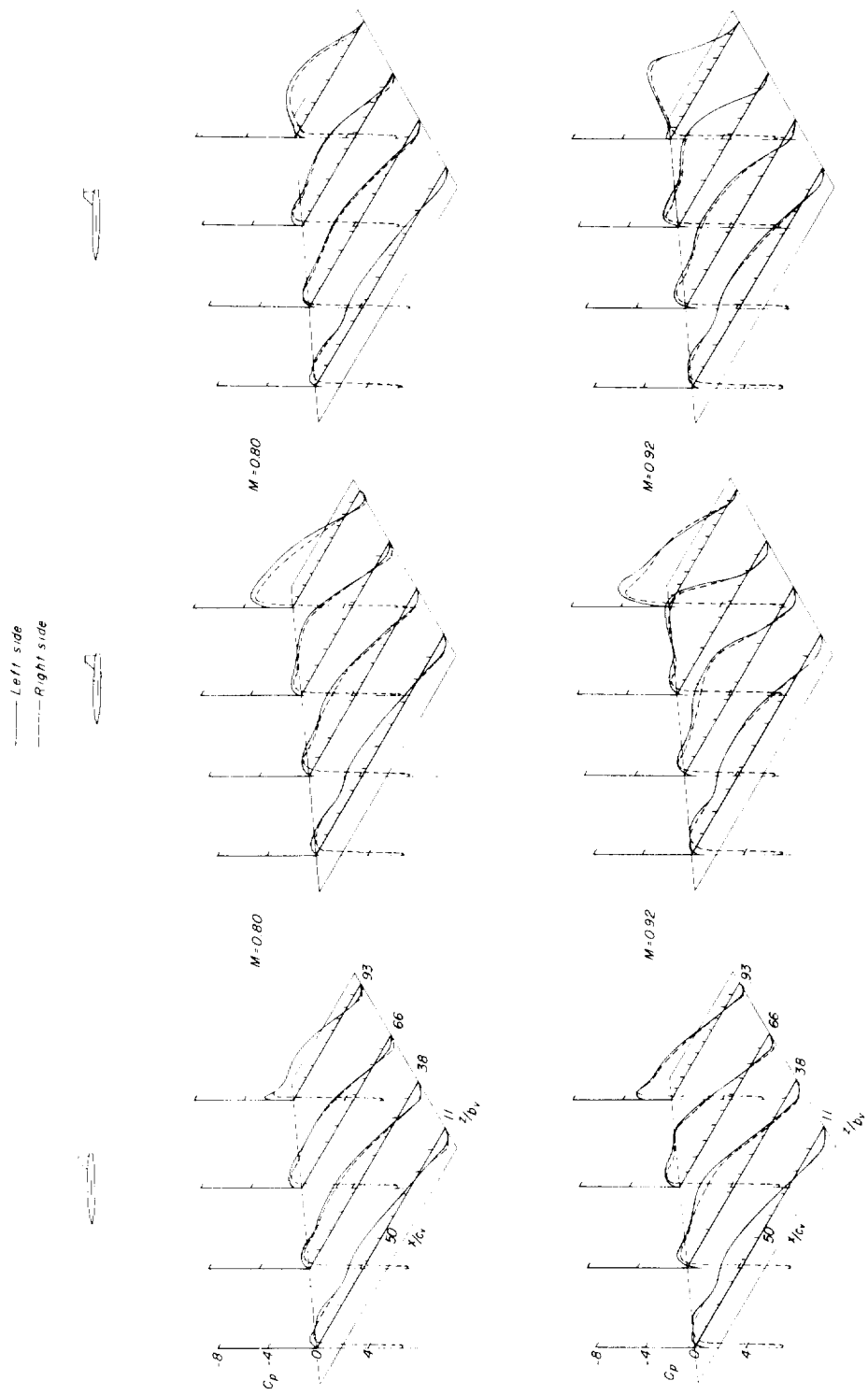


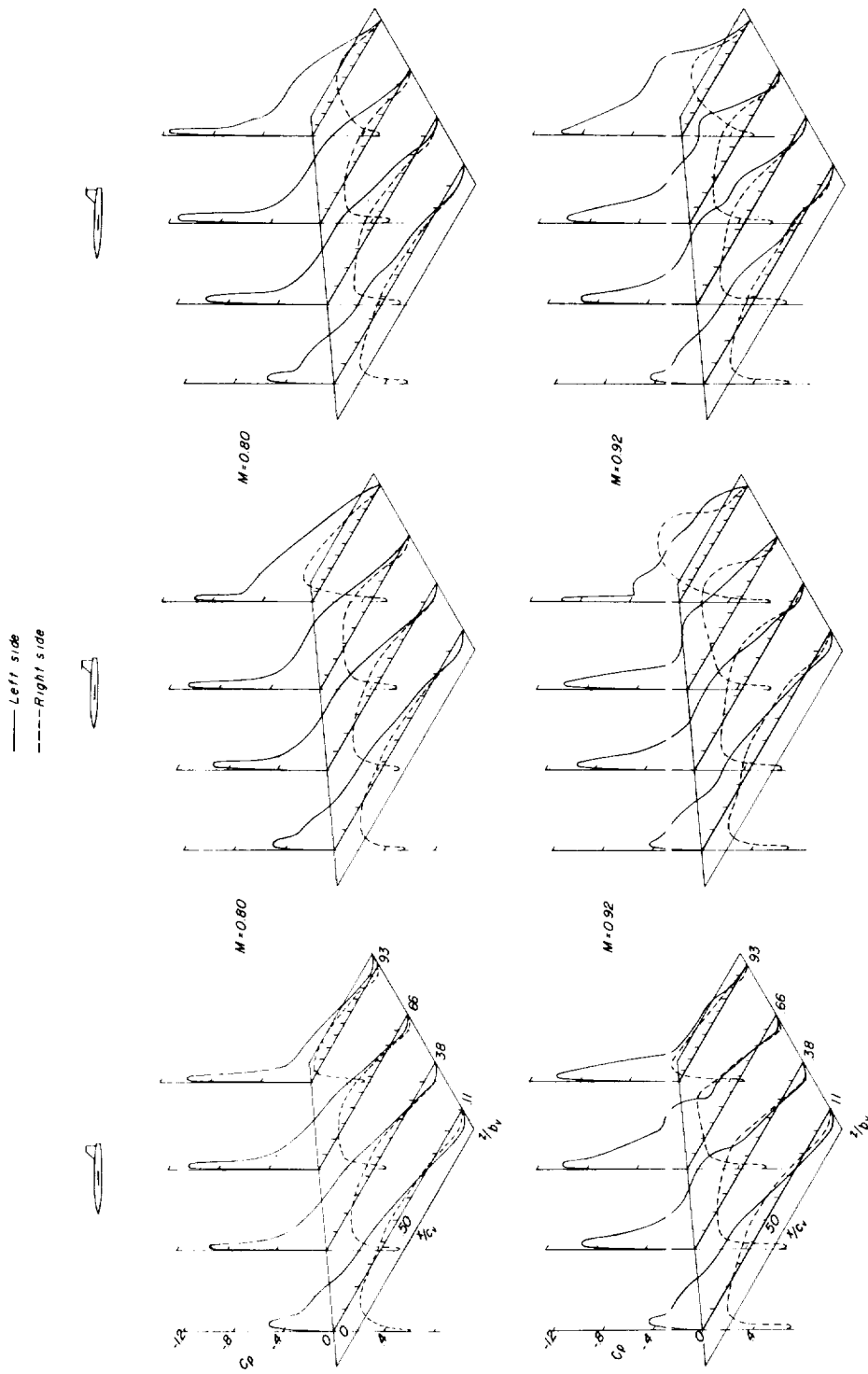
Figure 3.- Variation of model lift coefficient with angle of attack. Horizontal tail off. (Data from ref. 10.)





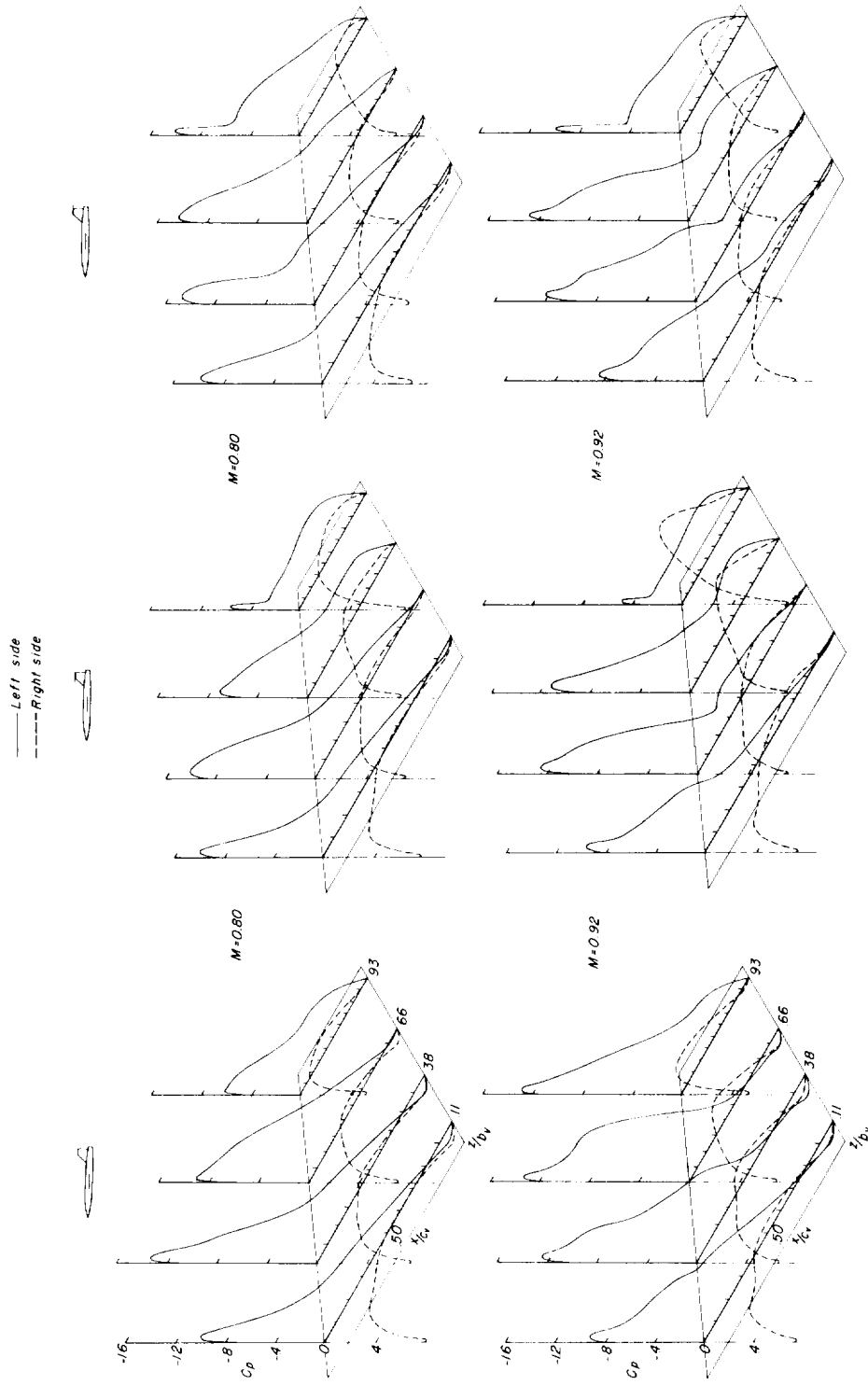
(a)  $\beta = 0^\circ$ .

Figure 4.- Effect of horizontal-tail position on vertical-tail pressure distribution.  $i_t = 0^\circ$ ;  
 $\alpha = 0^\circ$ .



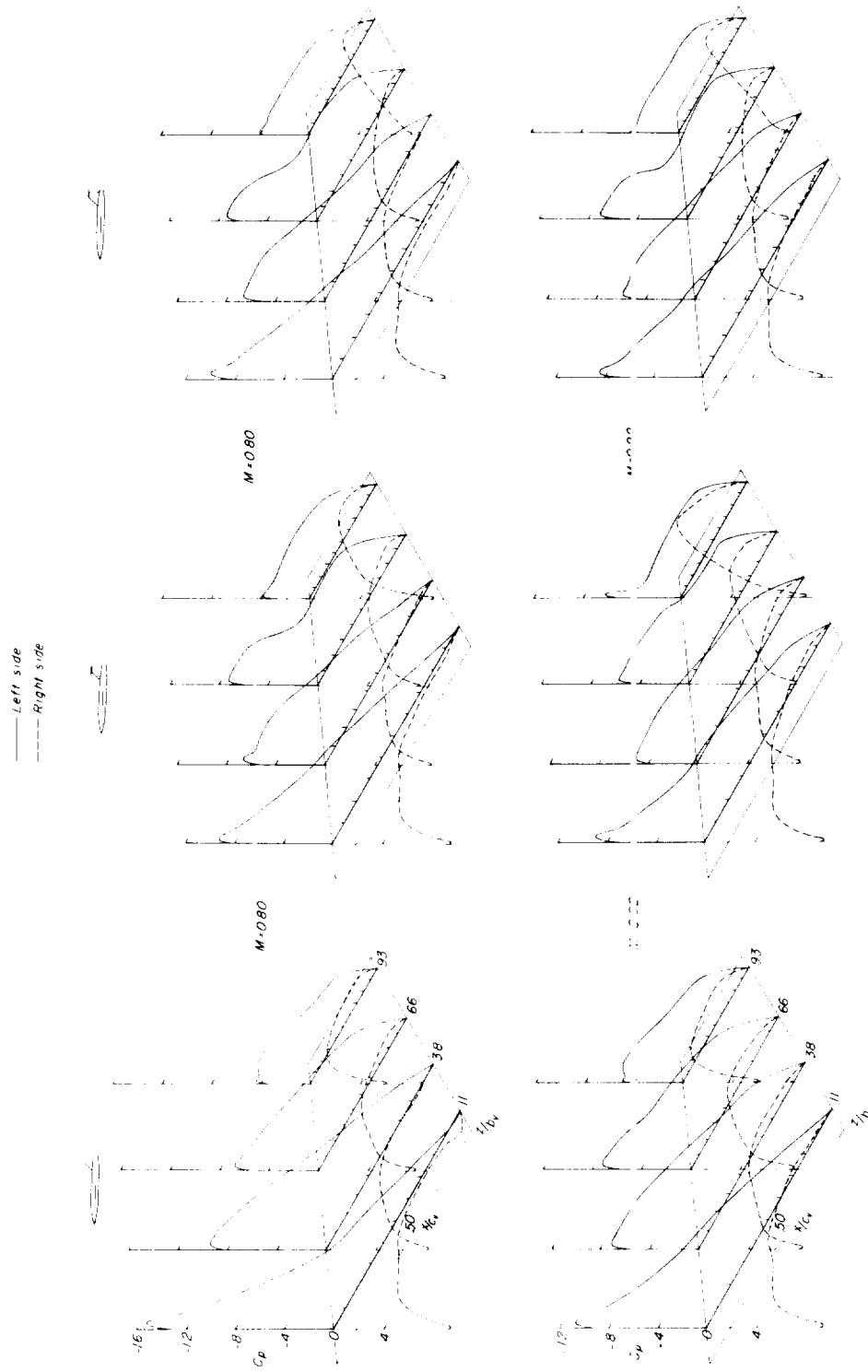
(b)  $\beta = 3.9^\circ$ .

Figure 4.- Continued.



(c)  $\beta = 7.8^\circ$ .

Figure 4.- Continued.



(d)  $\beta = 12.7^\circ$ .  
 Figure 4.- Concluded.

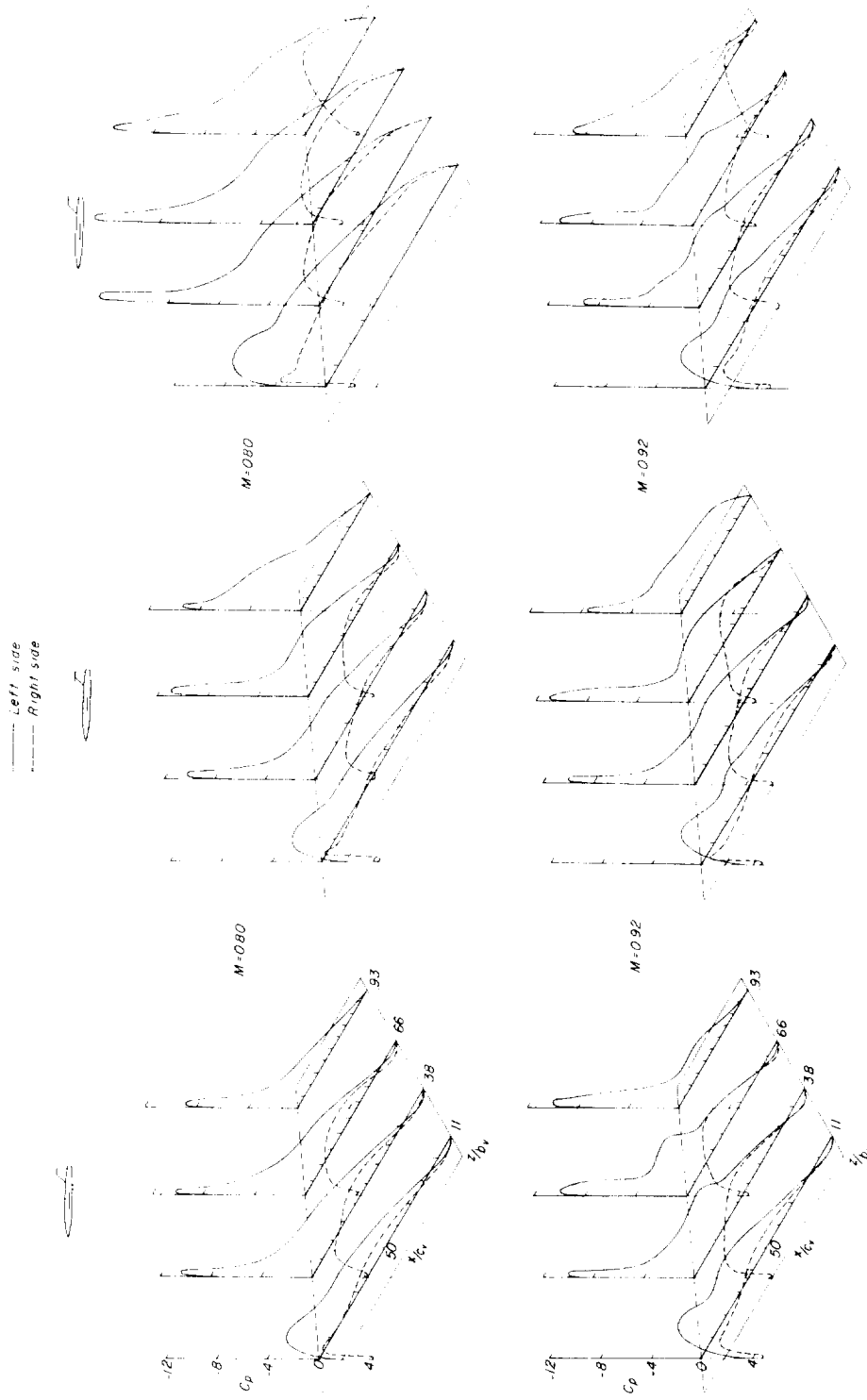
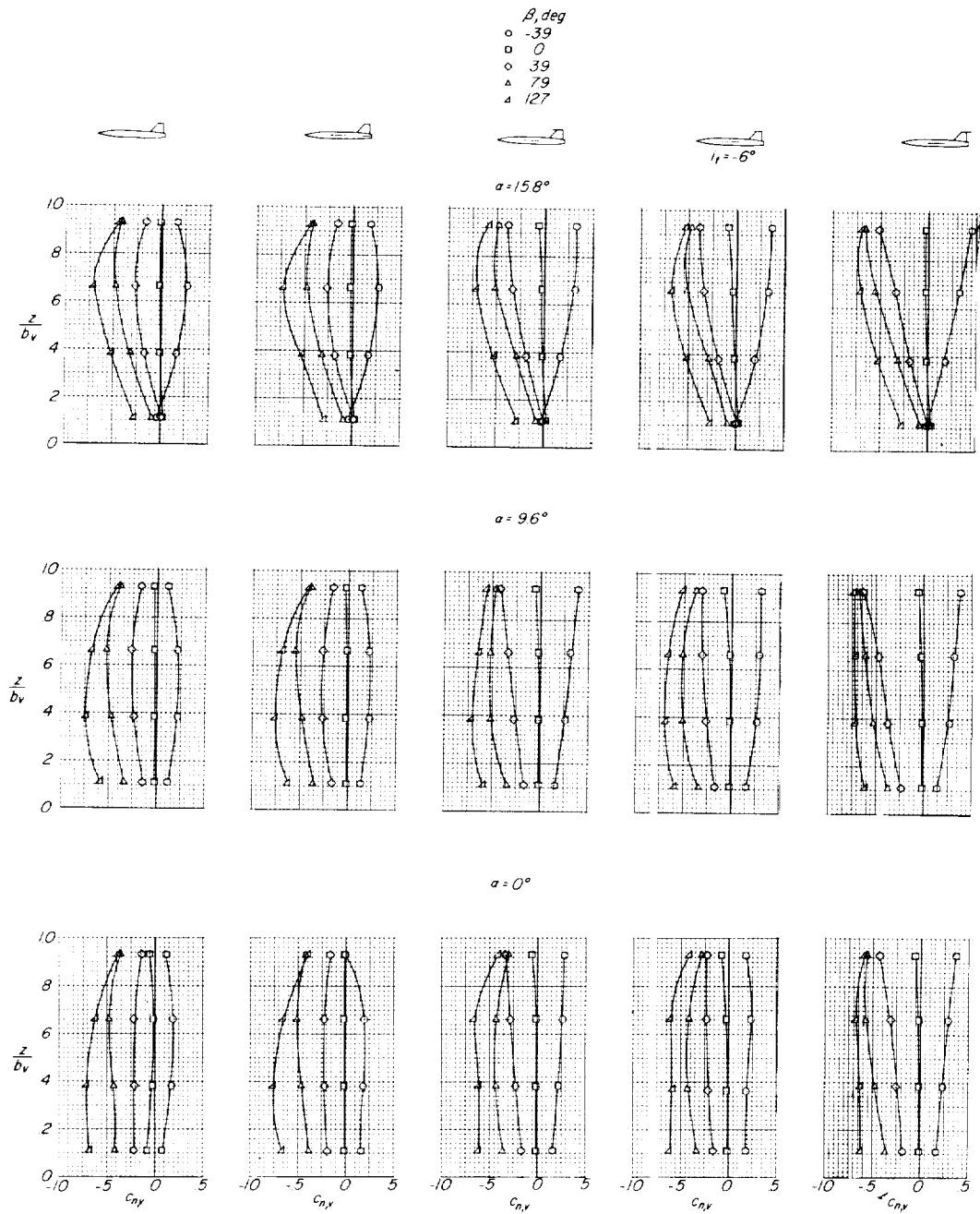
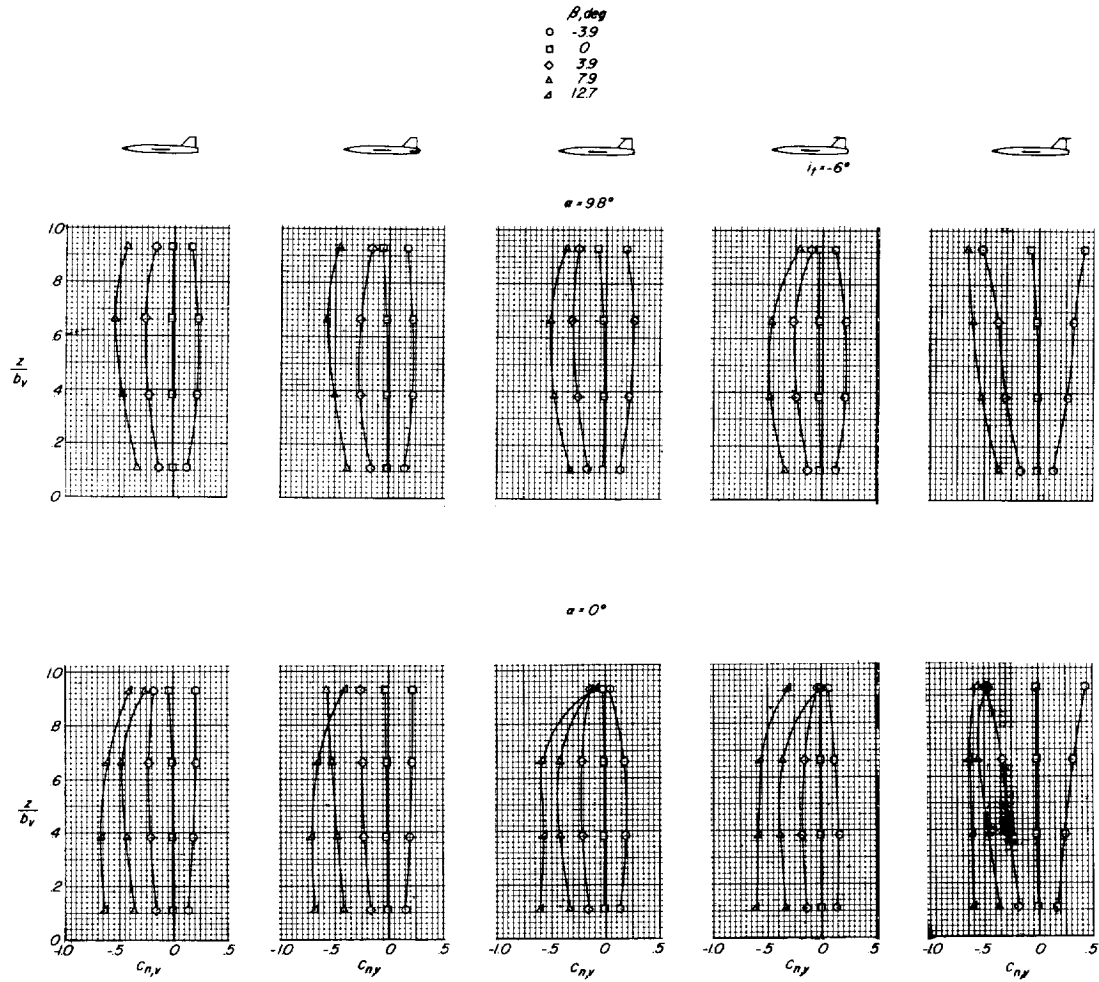


Figure 5.- Effect of horizontal-tail position on vertical-tail pressure distribution.  $i_t = 0^\circ$ ;  
 $\alpha = 7.6^\circ$ ;  $\beta = 3.9^\circ$ .



(a)  $M = 0.80$ .

Figure 6.- Effect of sideslip angle on the spanwise variation of vertical-tail section normal-force coefficient.  $i_t = 0^\circ$  (except where noted).



(b)  $M = 0.92$ .

Figure 6.- Concluded.

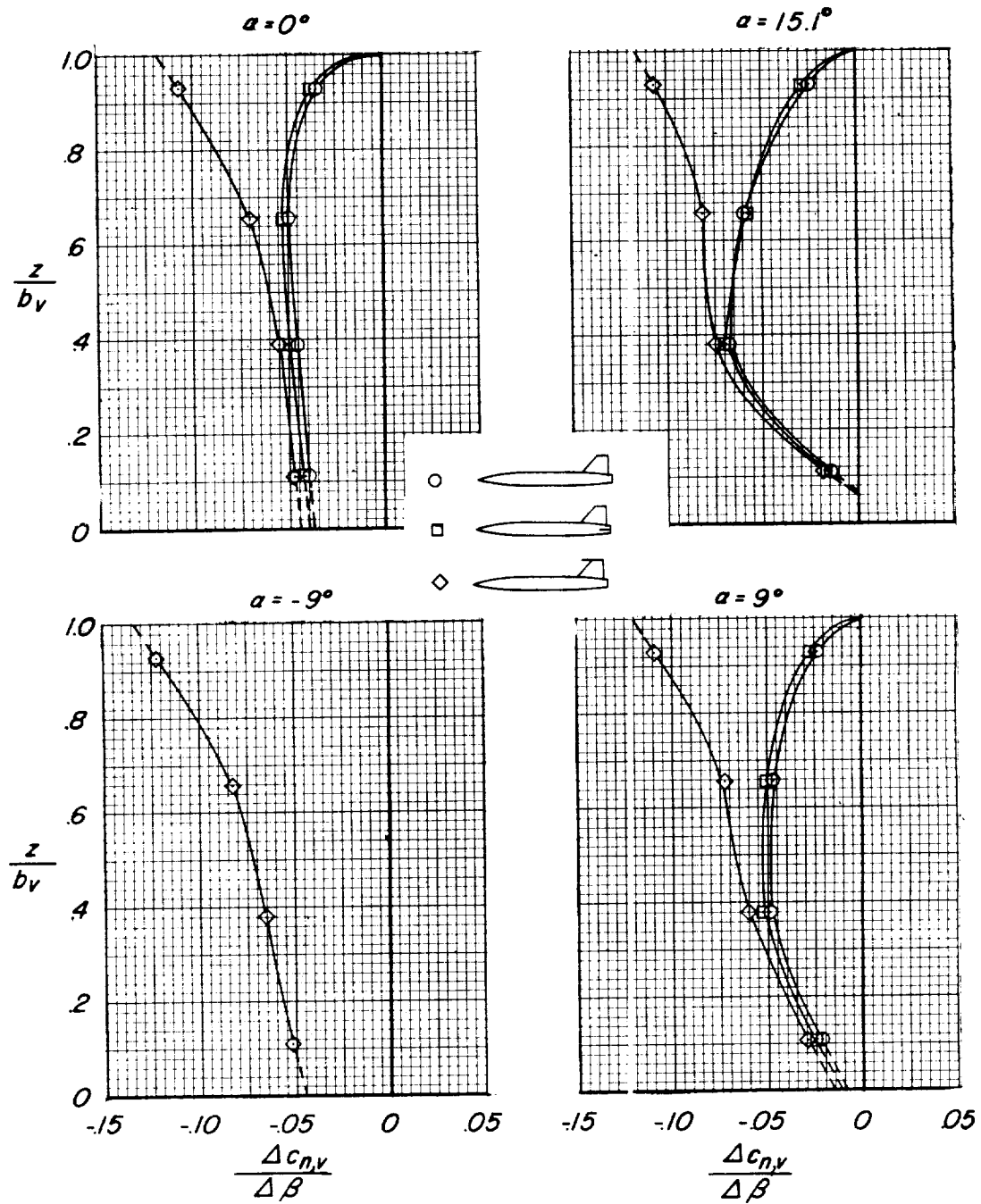
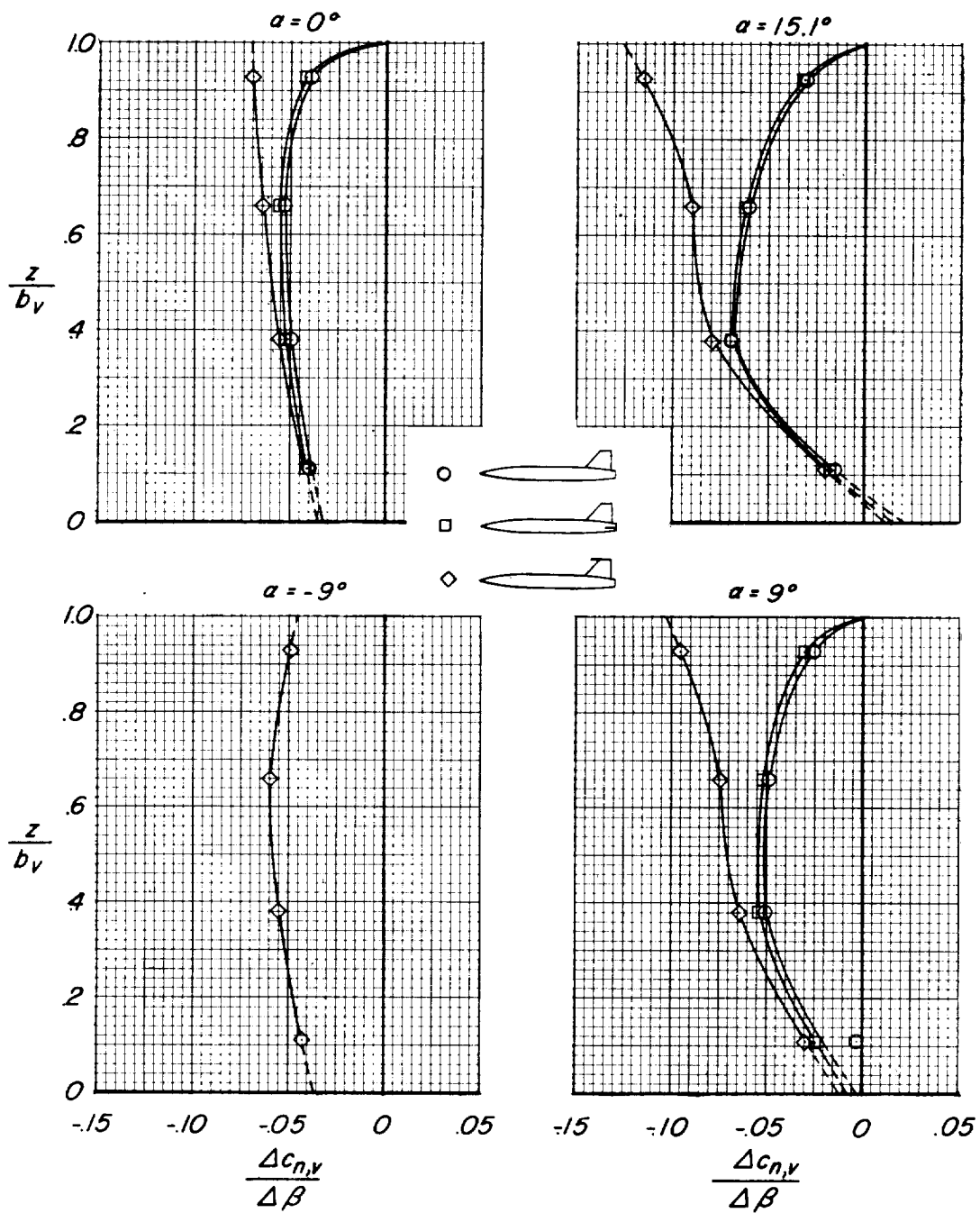
(a)  $M = 0.60$ .

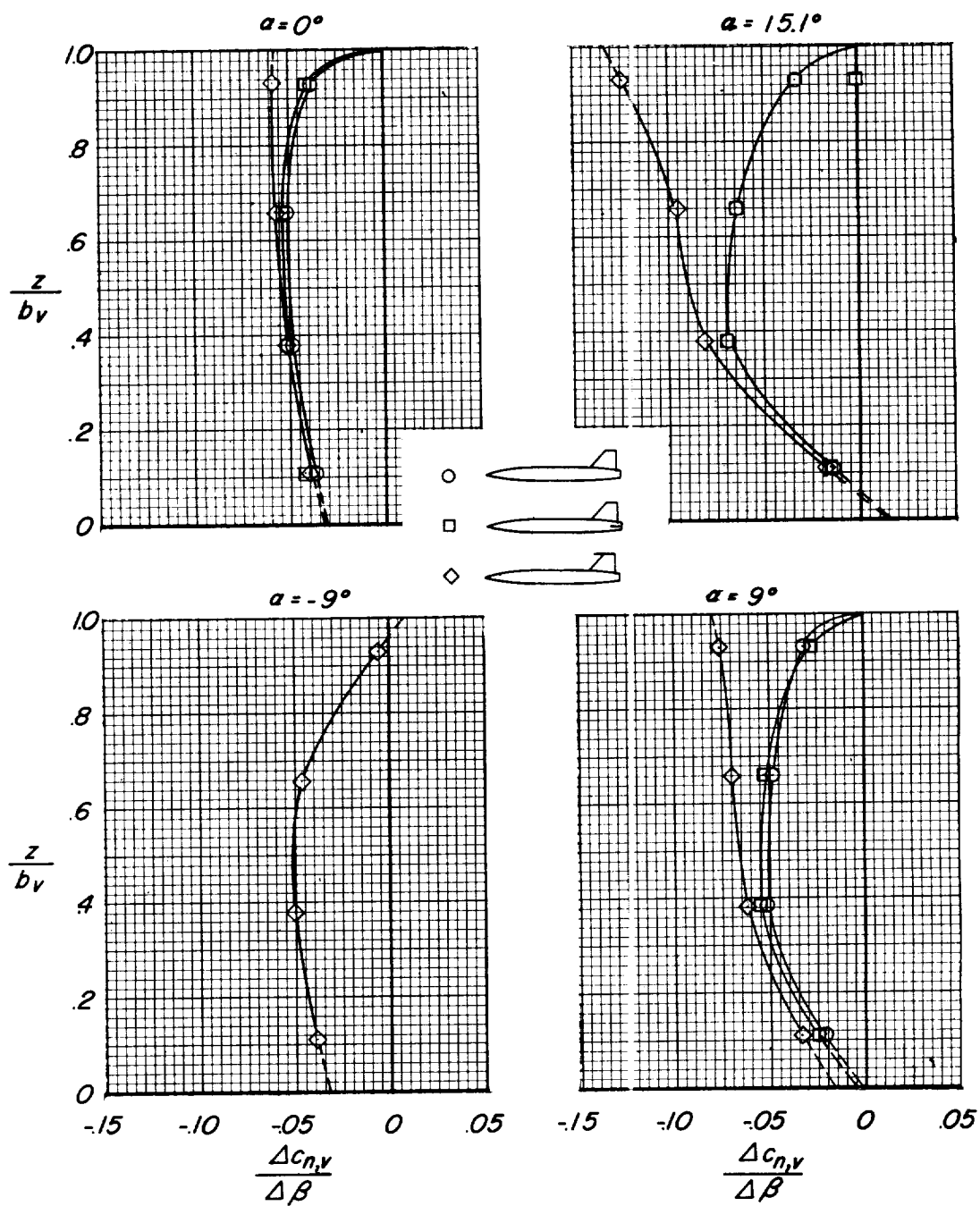
Figure 7.- Effect of horizontal-tail position on the spanwise variation of vertical-tail section normal-force coefficient per degree of sideslip. Wing-off configuration;  $\Delta \beta \approx 3^\circ$ ;  $i_t = 0^\circ$ .





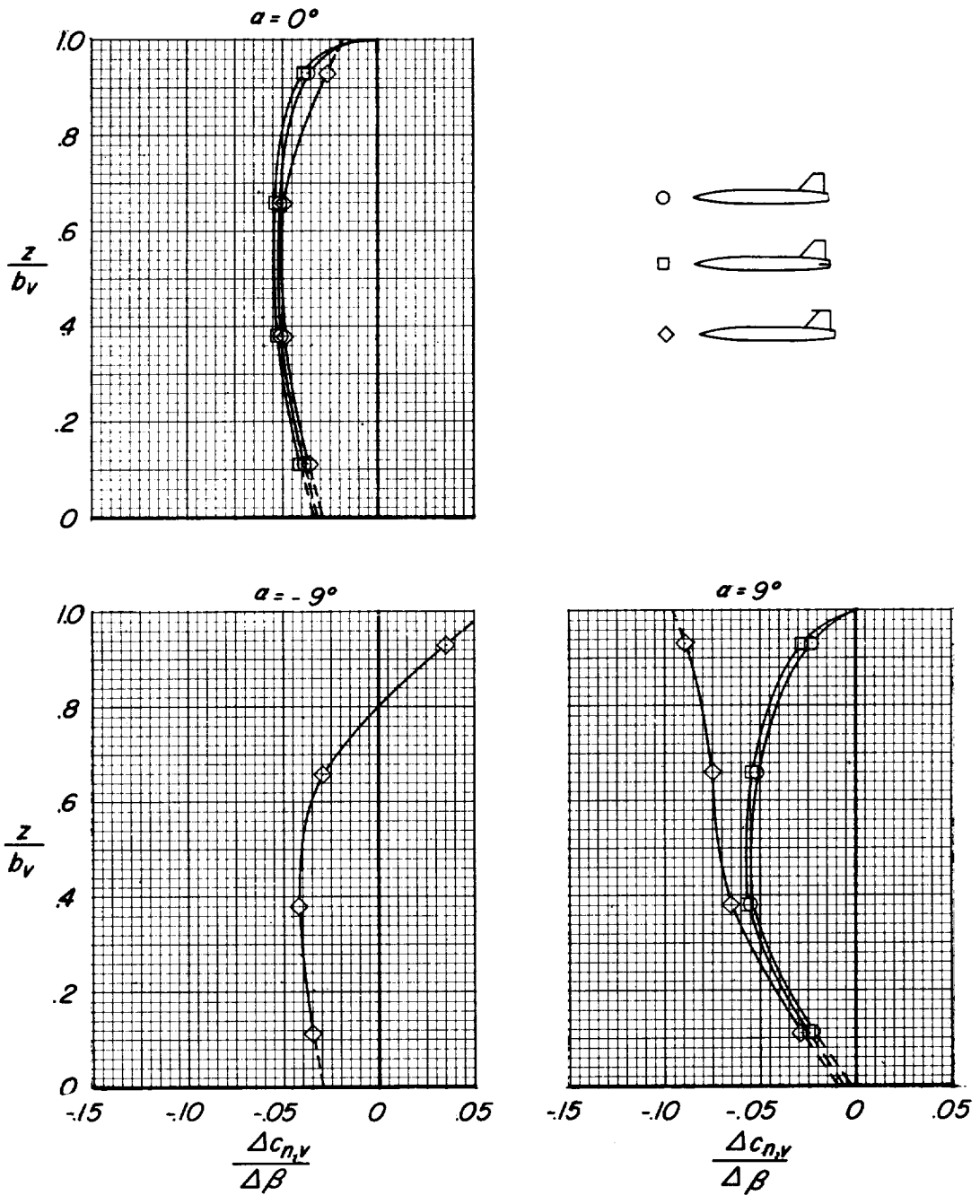
(b)  $M = 0.80$ .

Figure 7.- Continued.



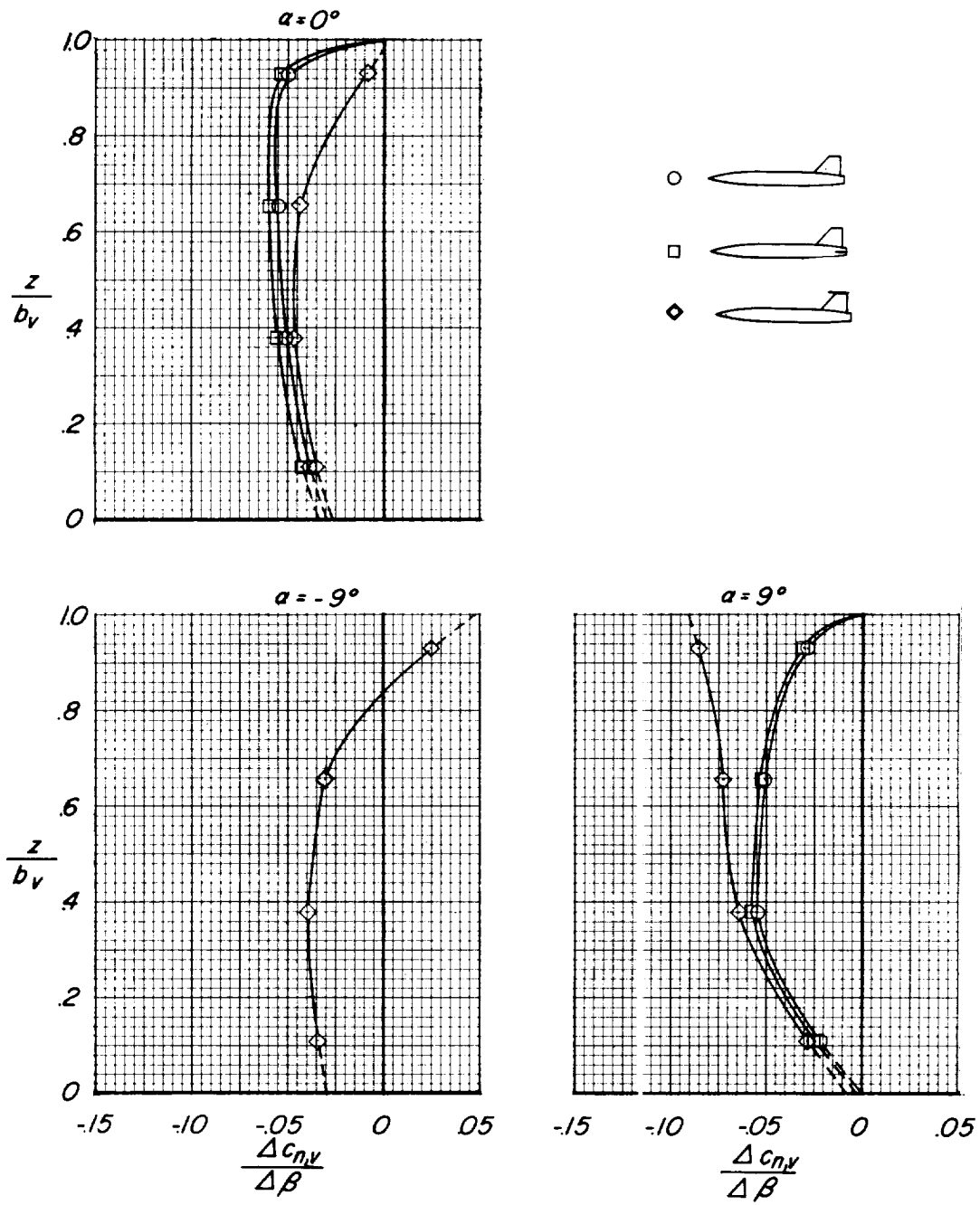
(c)  $M = 0.85$ .

Figure 7.- Contin.ed.



(d)  $M = 0.90$ .

Figure 7.- Continued.



(e)  $M = 0.92$ .

Figure 7.- Concluded.

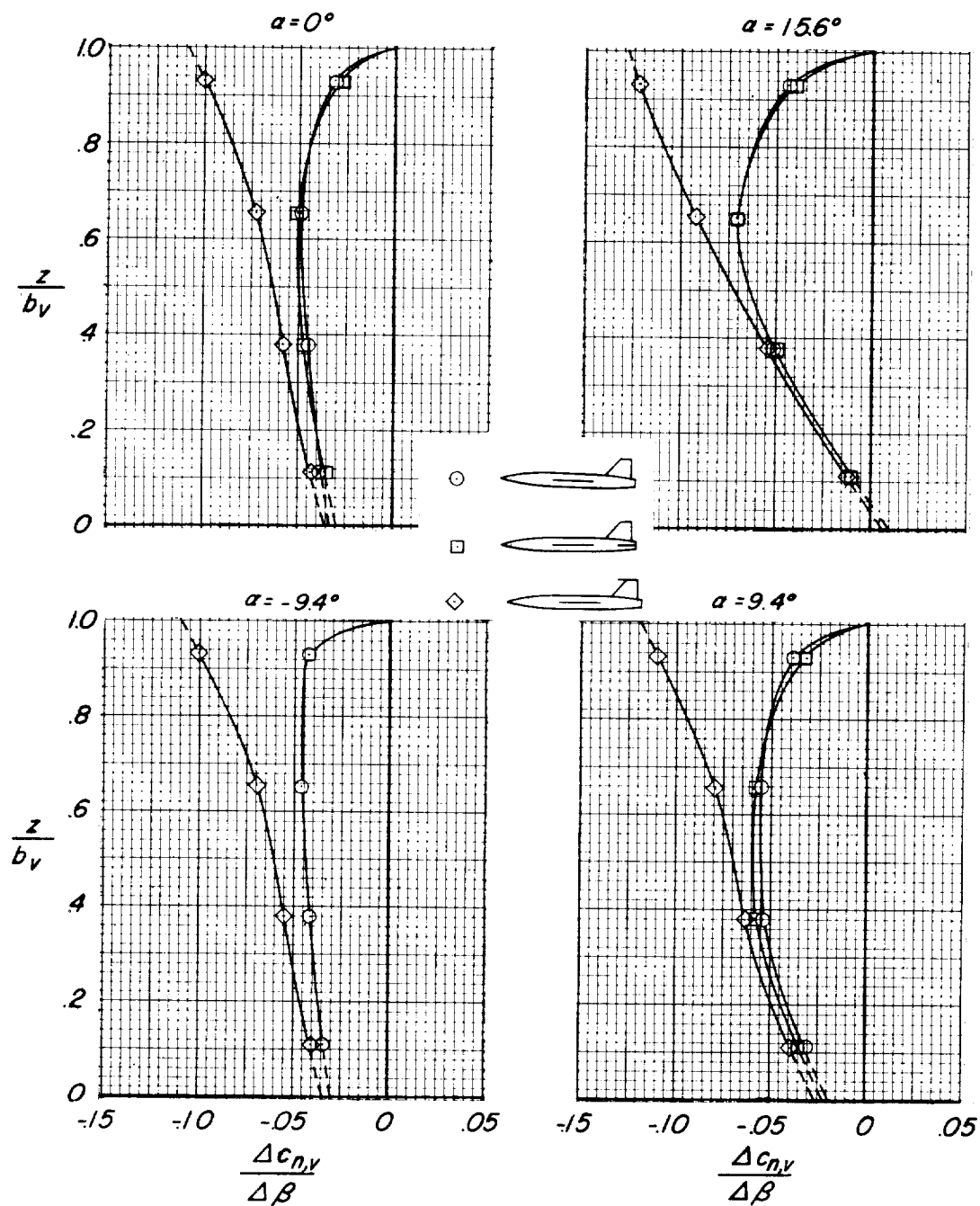
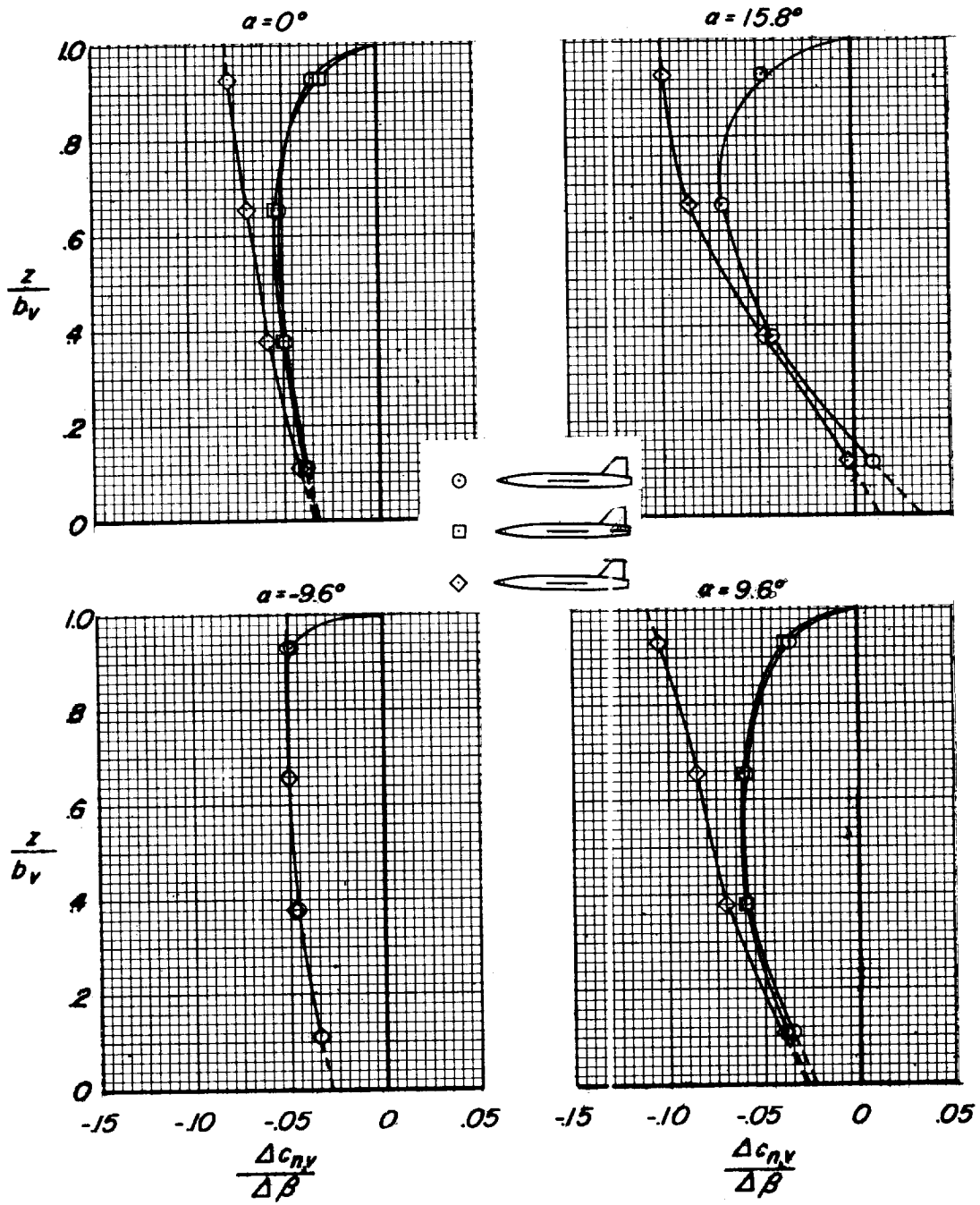
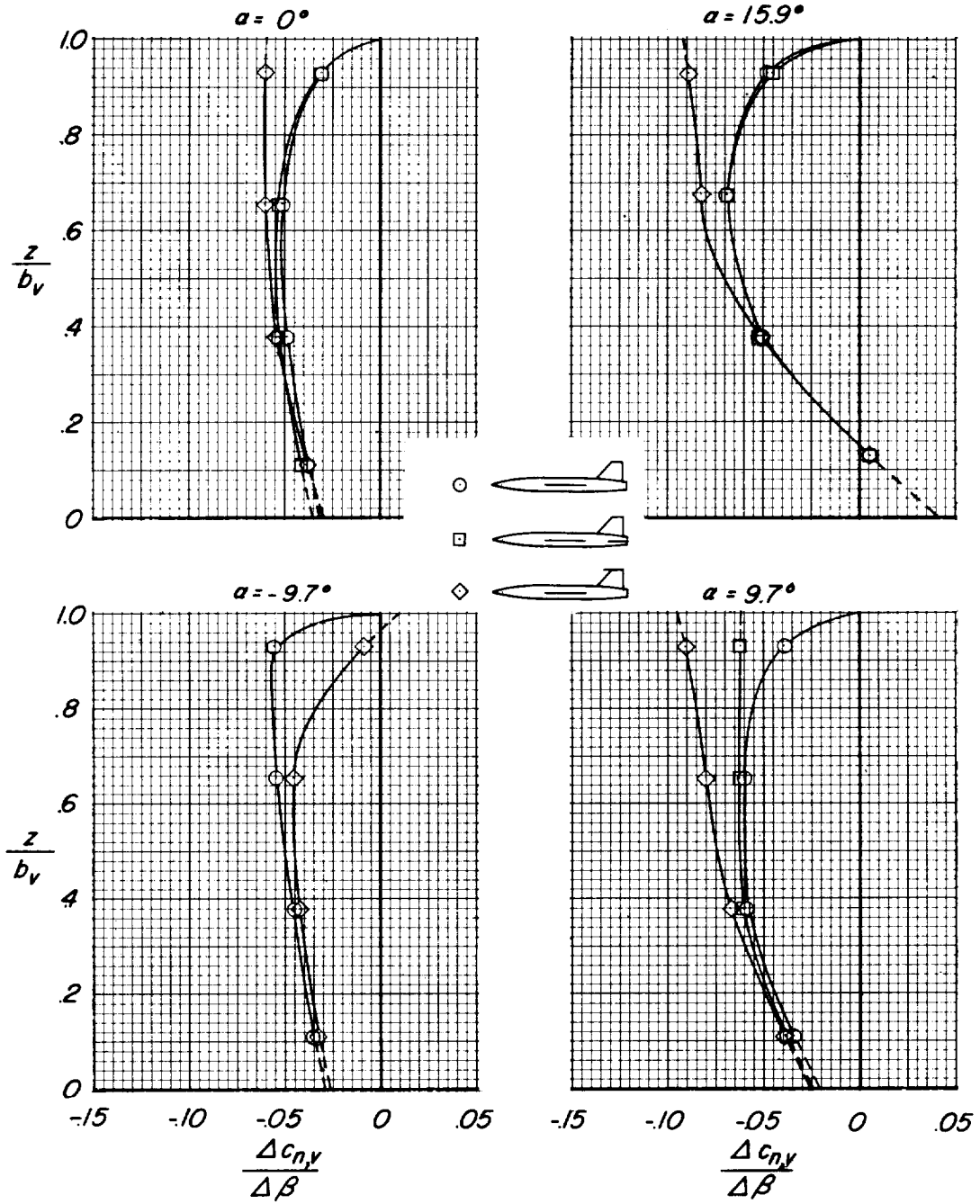
(a)  $M = 0.60$ .

Figure 8.- Effect of horizontal-tail position on the spanwise variation of vertical-tail section normal-force coefficient per degree of sideslip.  $\Delta \beta \approx 8^\circ$ ;  $i_t = 0^\circ$ .



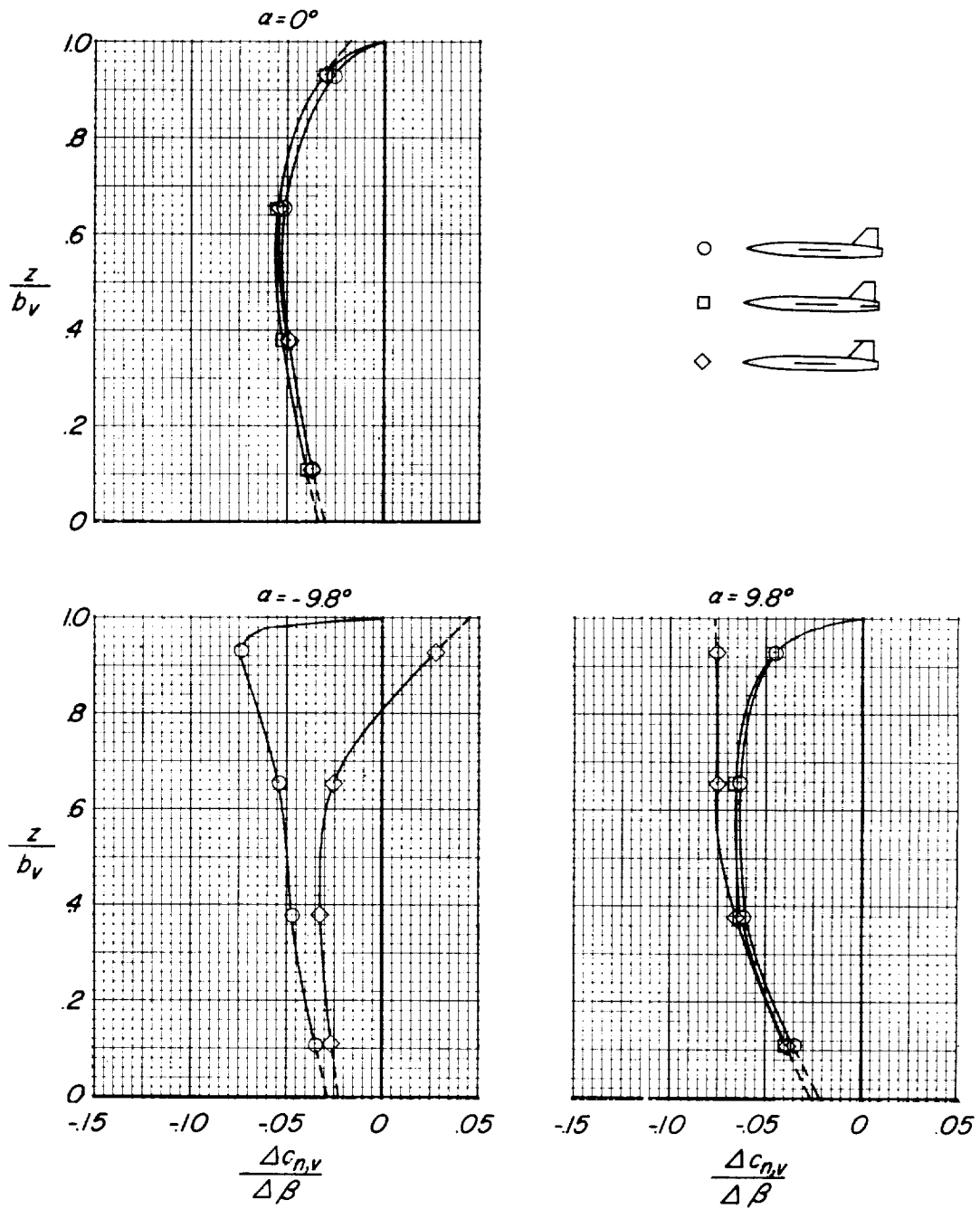
(b)  $M = 0.80$ .

Figure 8.- Continued.



(c)  $M = 0.85$ .

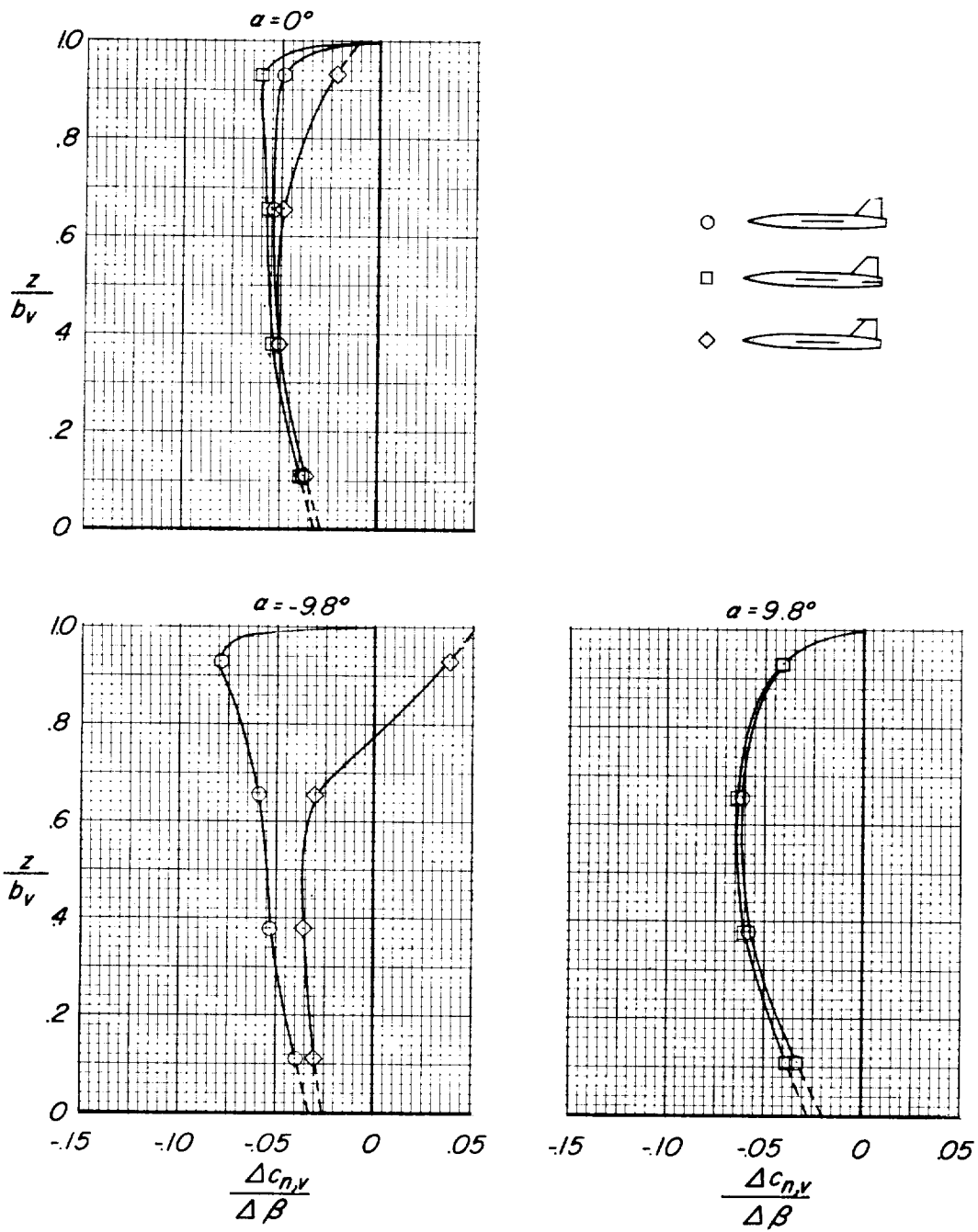
Figure 8.- Continued.



(d)  $M = 0.90$ .

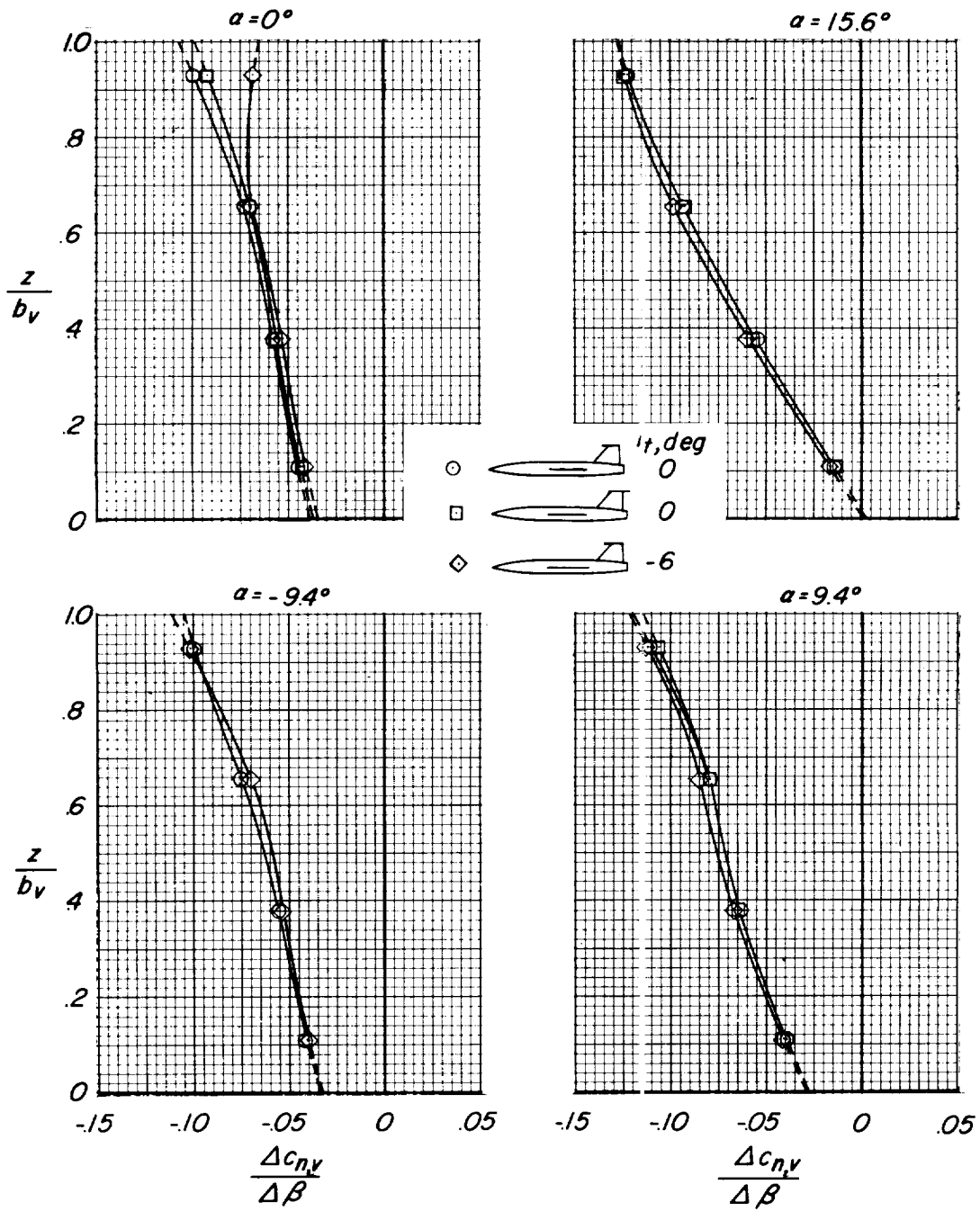
Figure 8.- Continued.





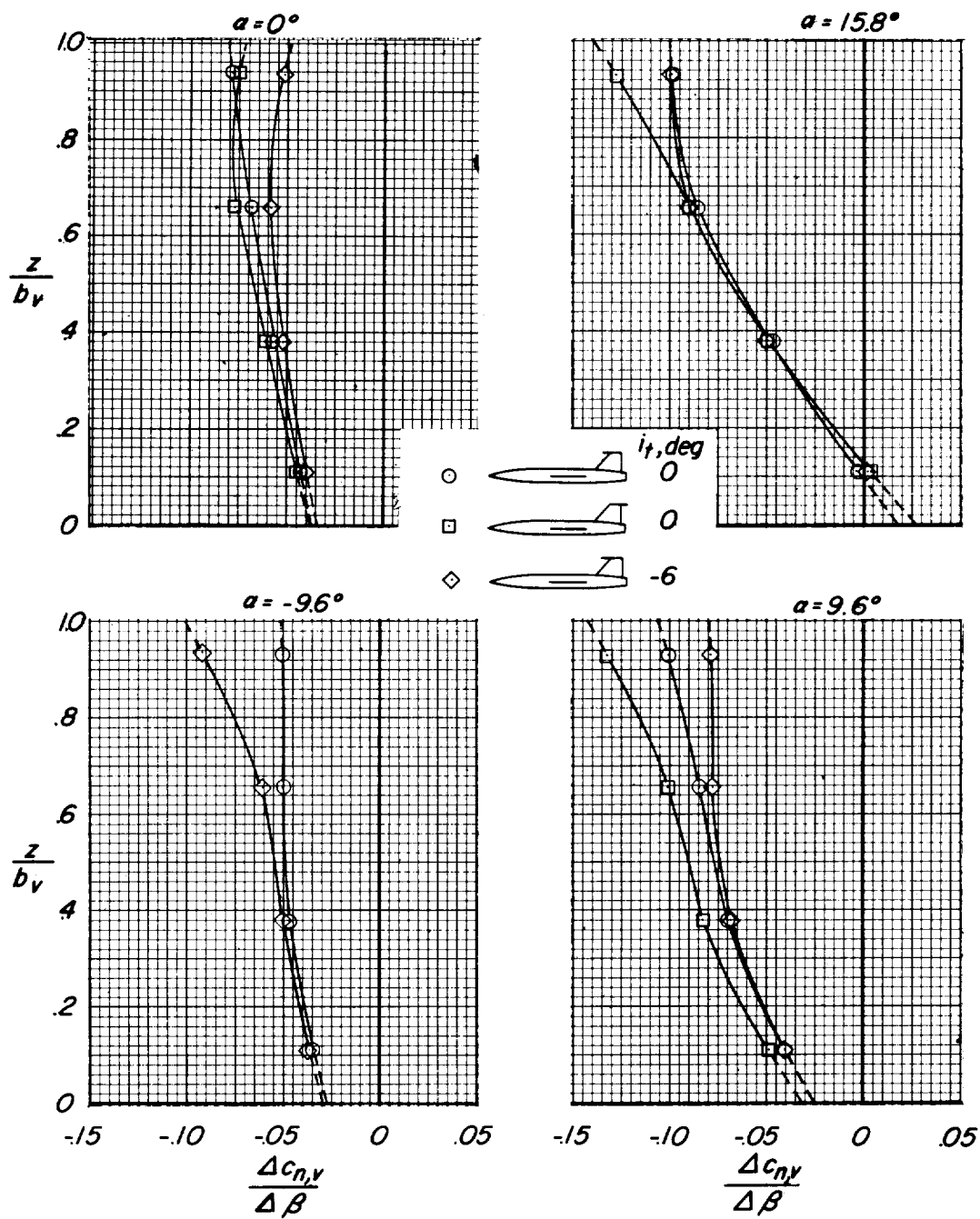
(e)  $M = 0.92$ .

Figure 8.- Concluded.



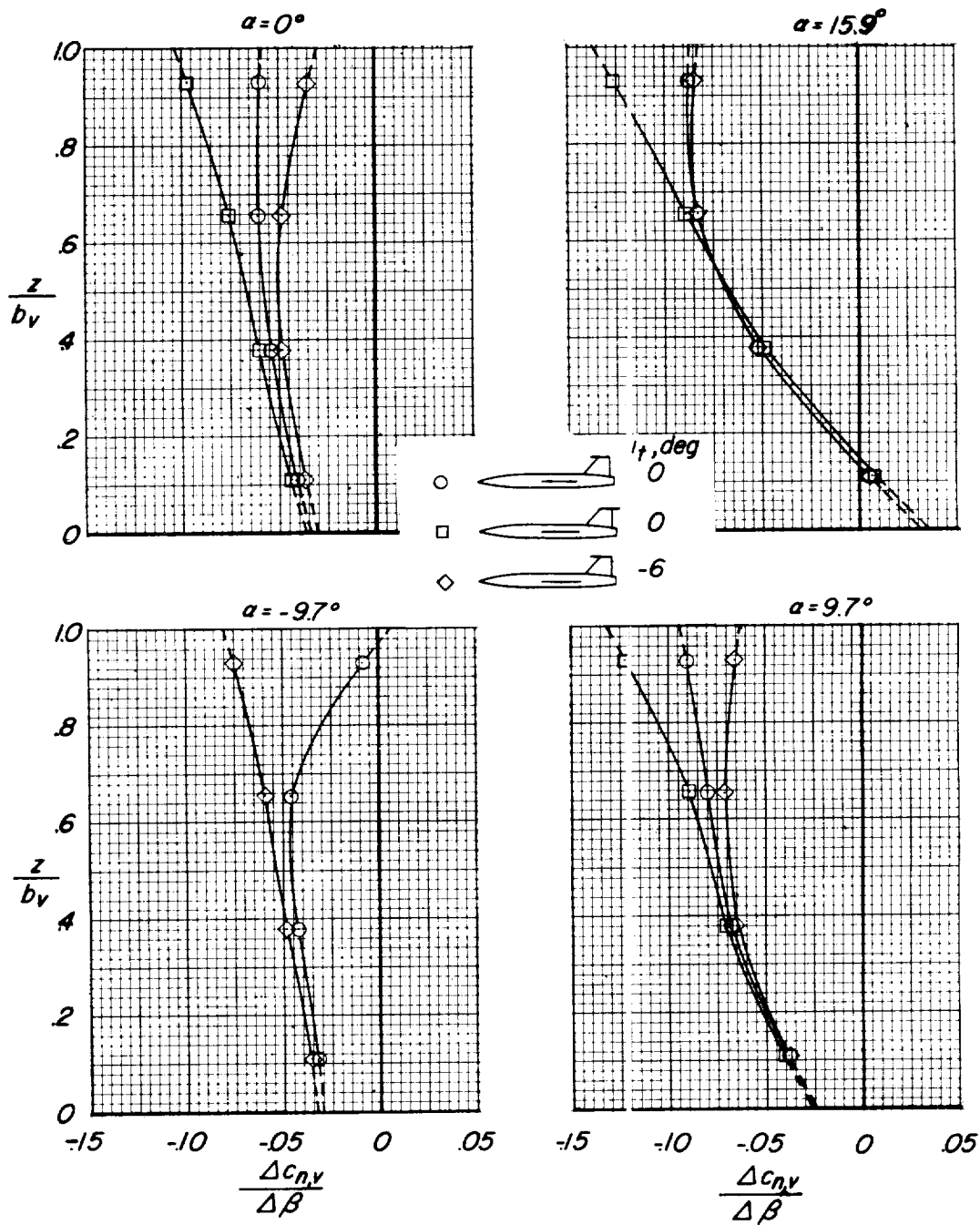
(a)  $M = 0.60$ .

Figure 9.- Effect of horizontal-tail position and incidence on the spanwise variation of vertical-tail section normal-force coefficient per degree of sideslip.  $\Delta \beta \approx 8^\circ$ .



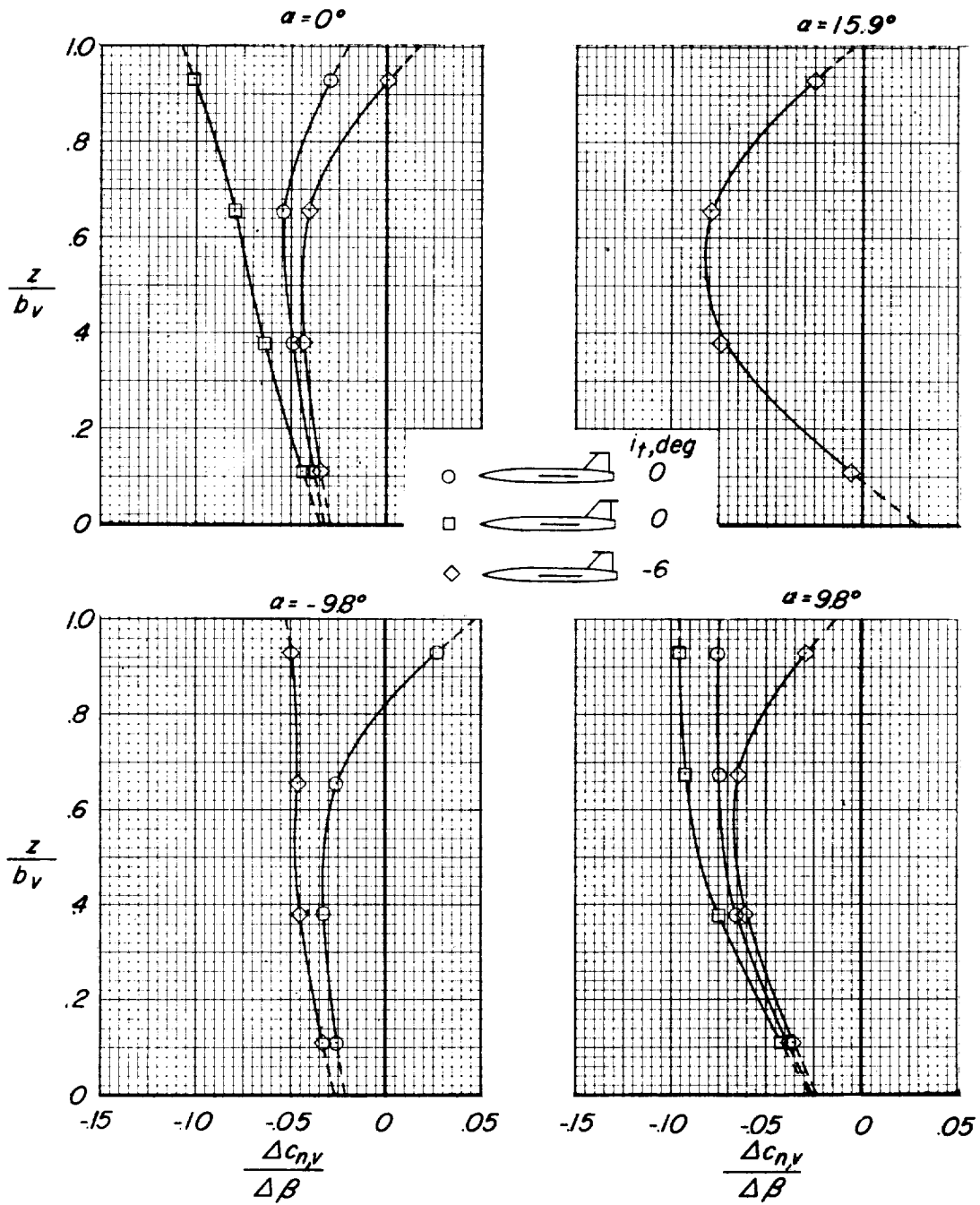
(b)  $M = 0.80$ .

Figure 9.- Continued.



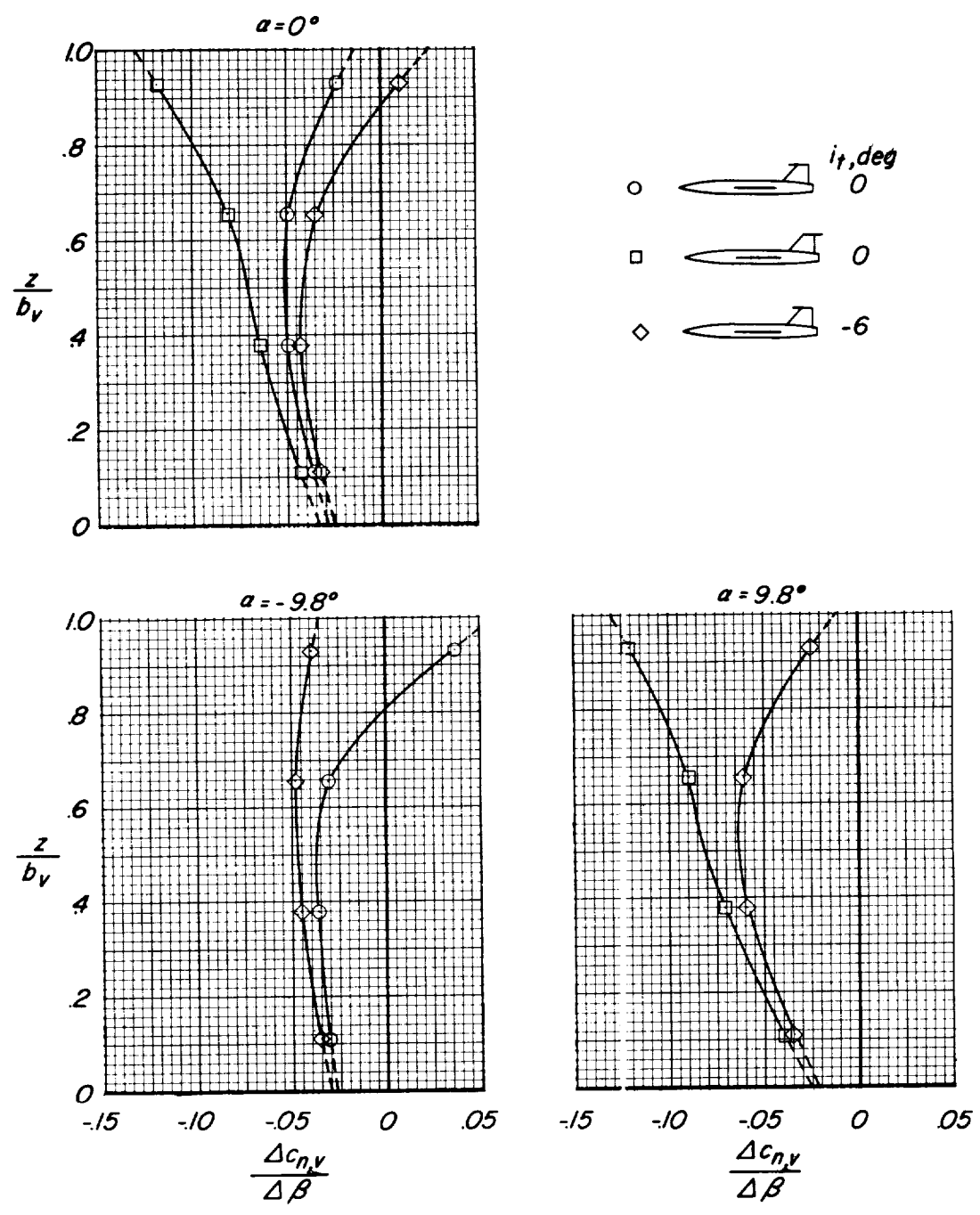
(c)  $M = 0.85$ .

Figure 9.- Continued.



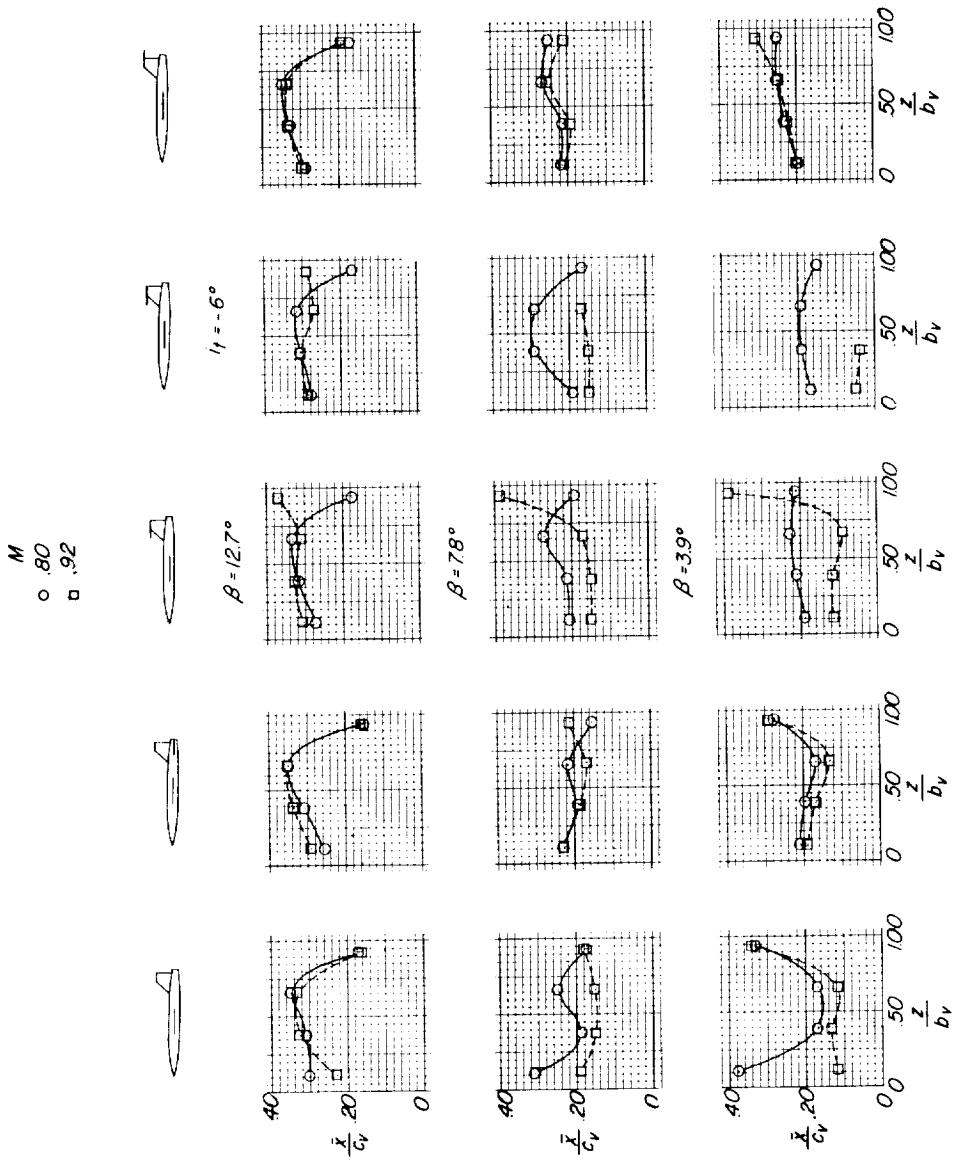
(d)  $M = 0.90$ .

Figure 9.- Continued.



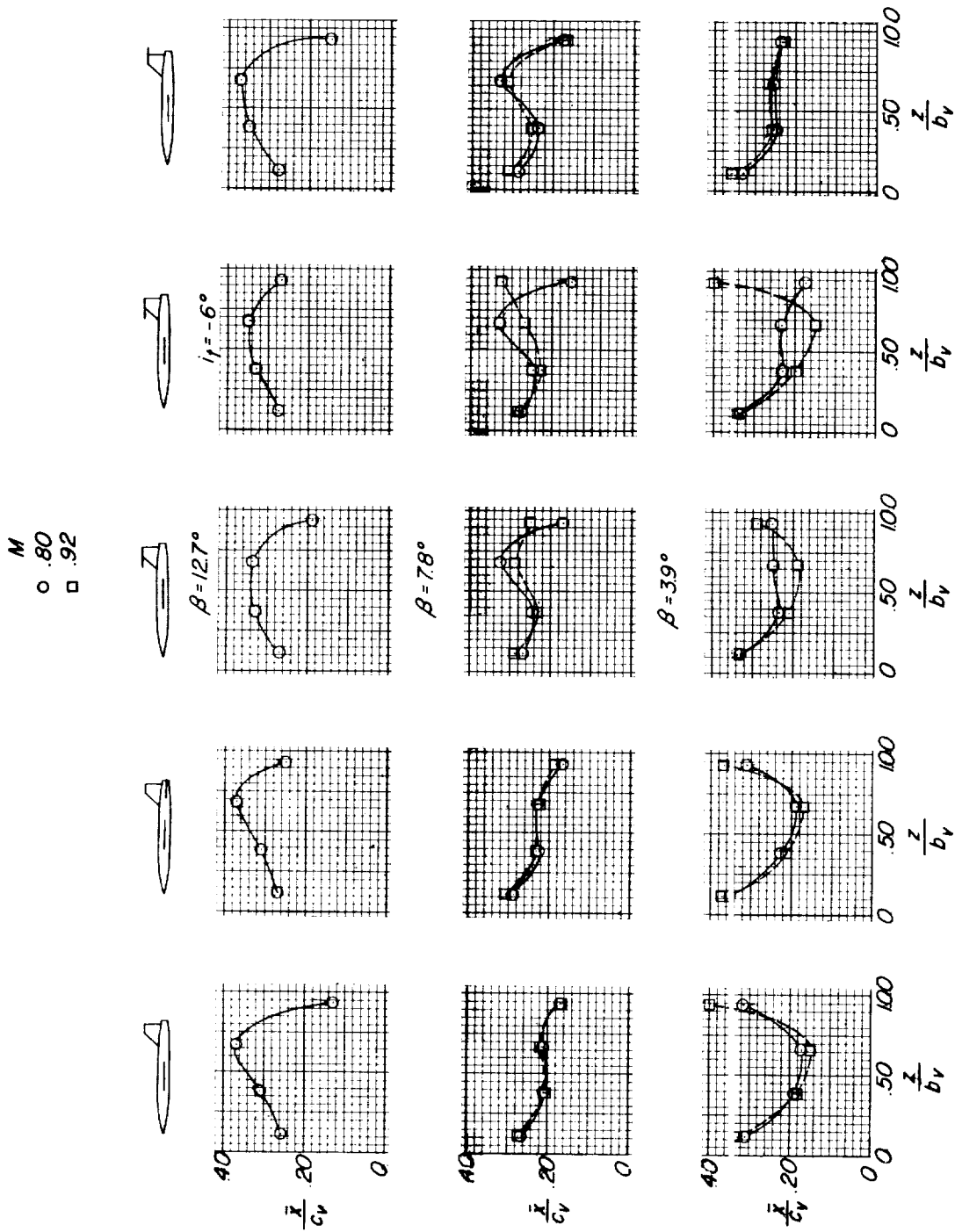
(e)  $M = 0.92$ .

Figure 9.- Concluded.



(a)  $\alpha = 0^\circ$ .

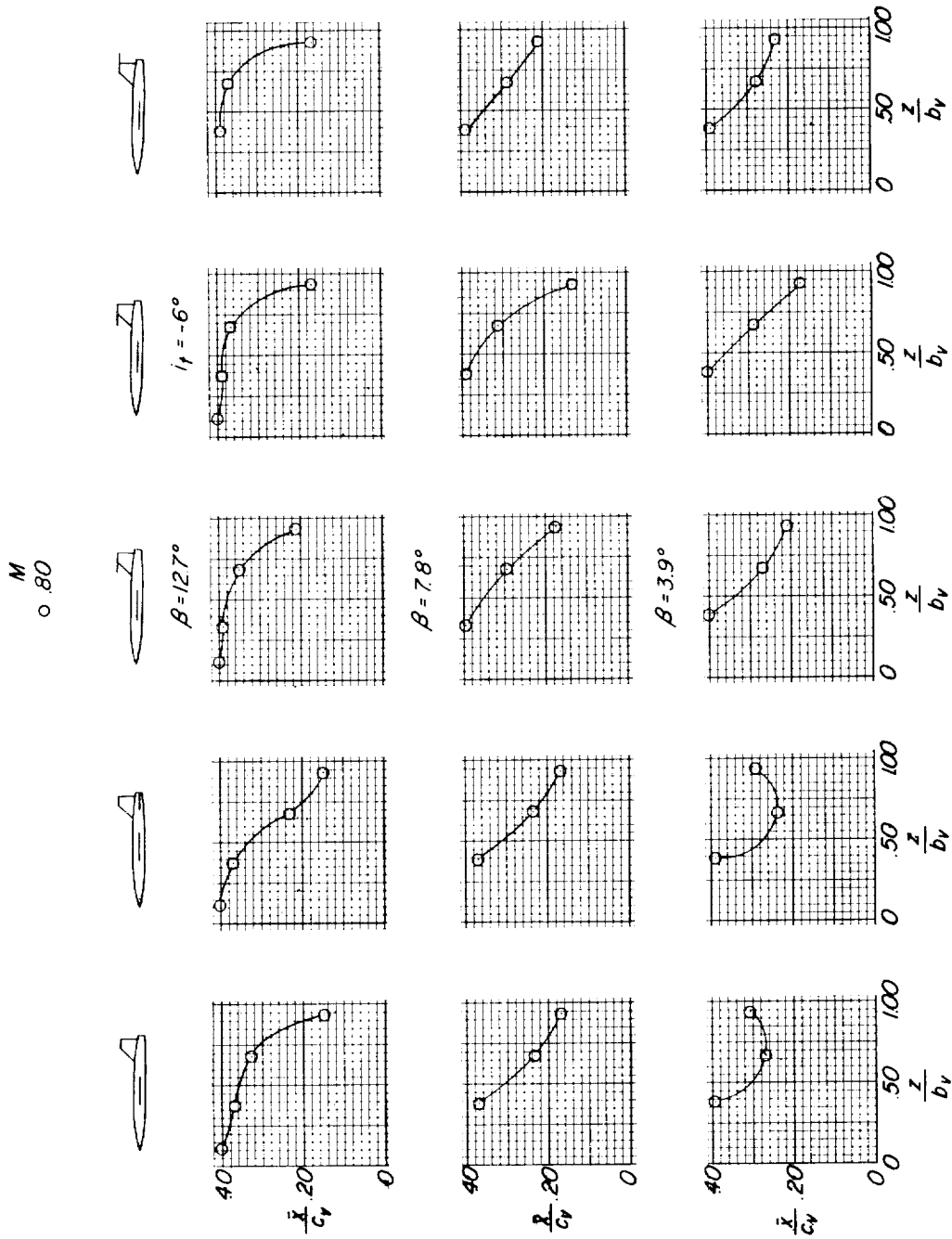
Figure 10.- Effect of horizontal-tail position and incidence on the chordwise location of vertical-tail local centers of pressure.  $i_t = 0^\circ$  (except where noted).



(b)  $\alpha \approx 9.6^\circ$ .

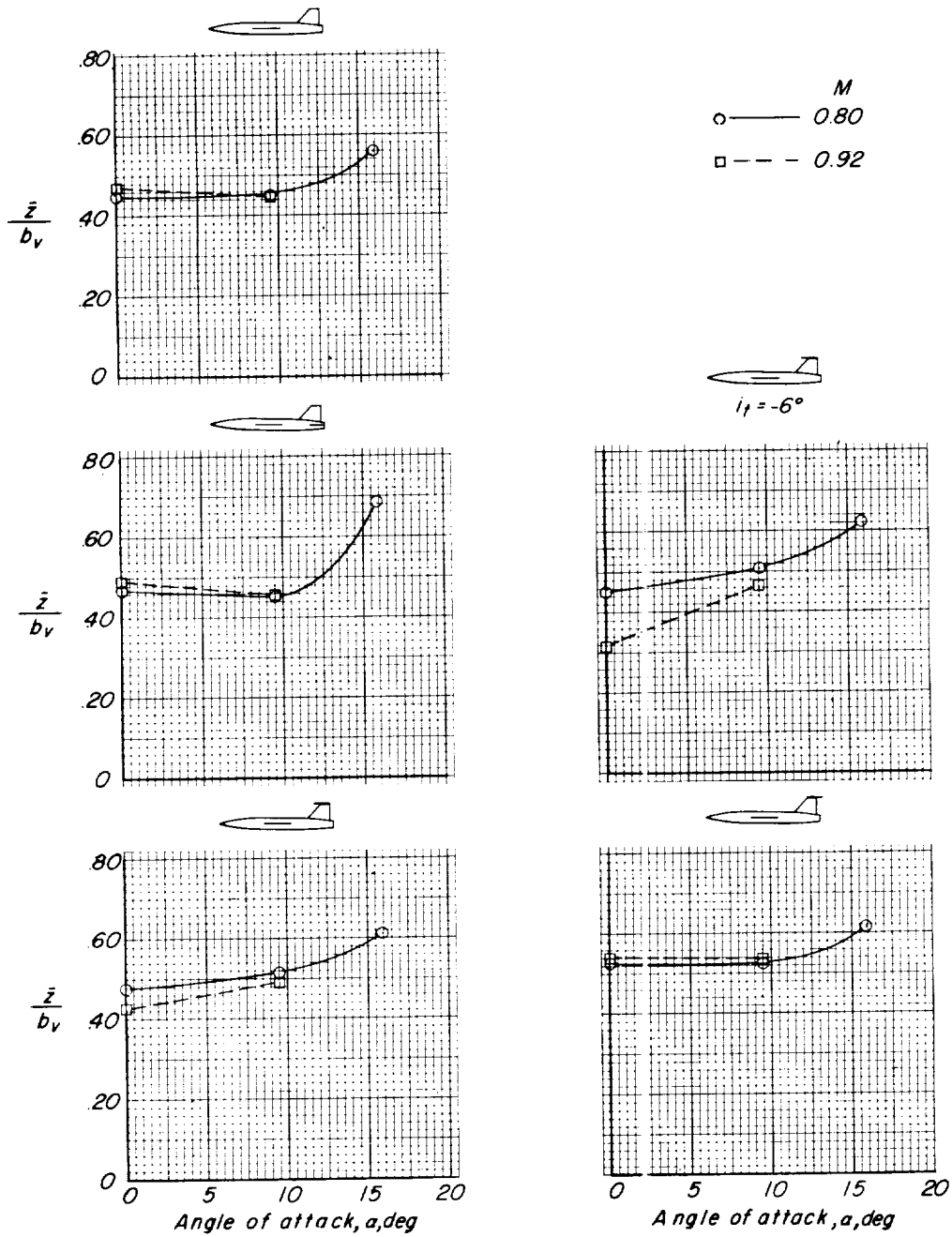
Figure 10.- Continued.





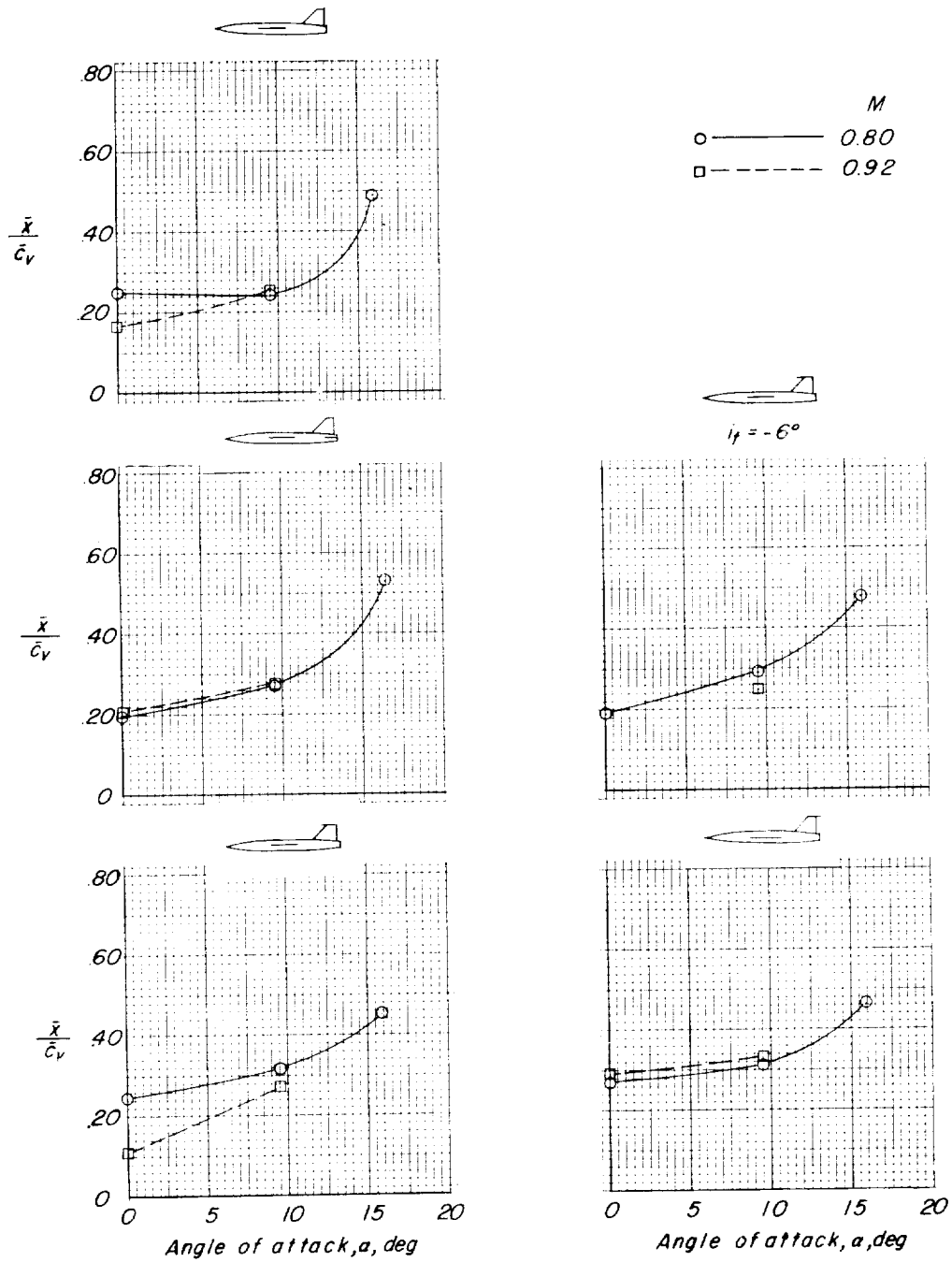
(c)  $\alpha \approx 15.8^\circ$ .

Figure 10.- Concluded.



(a) Spanwise variation.

Figure 11.- Effect of horizontal-tail position and incidence on the variation of the center of pressure of the total vertical-tail load with angle of attack.  $\Delta\beta \approx 8^\circ$ .



(b) Chordwise variation.

Figure 11.- Concluded.

