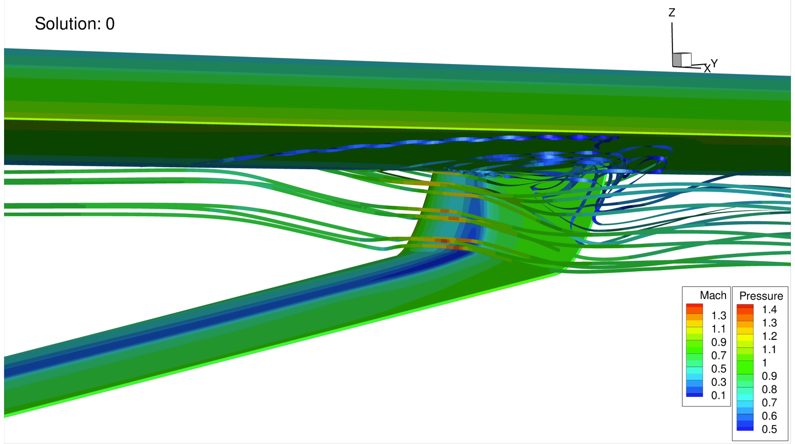
NASA Ames Research Center Contributions to the PADRI workshop



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NASA Ames Research Center

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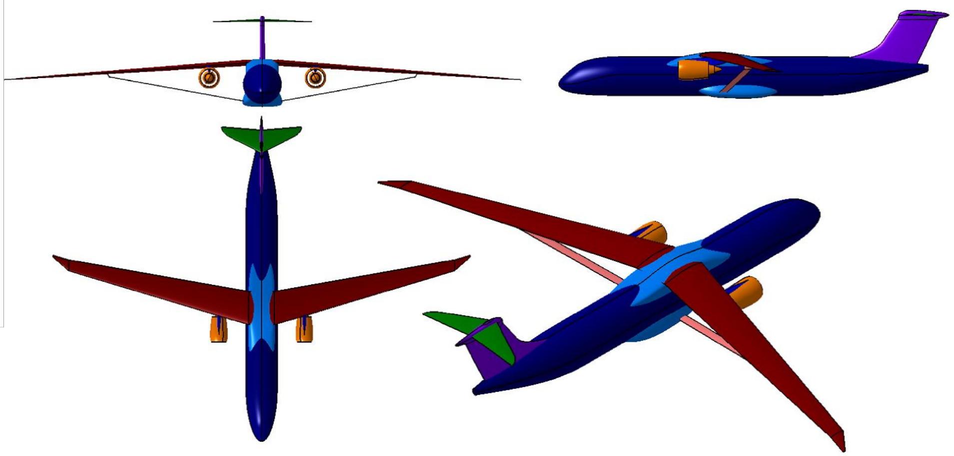
PADRI

- PADRI: A common platform for validation of aircraft drag reduction technologies
- Generic strut-braced wing configuration
- Slightly swept wing for low cruise Mach number (0.72)
- Simplified geometry without engines, empennage or flap-track fairings
- Significant wave-drag and flow separation at strut-wing intersection
- Focus of this workshop is to redesign the junction



MDO for Aircraft Configurations with High-fidelity (MACH)

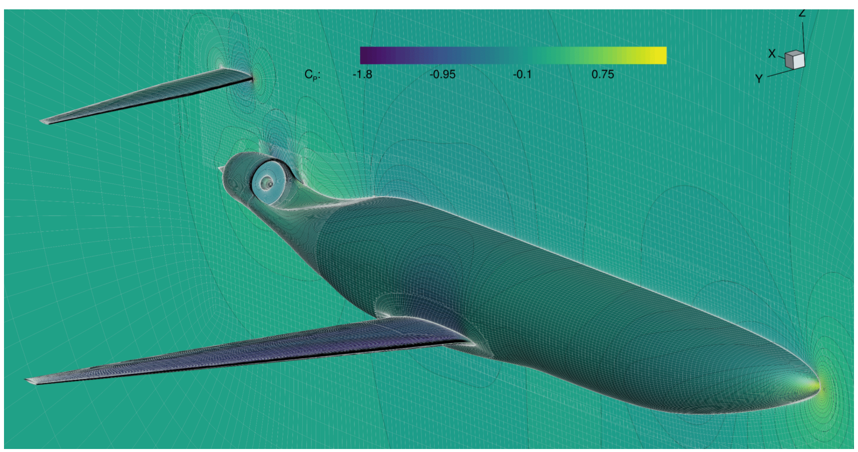
Python user script
Setup up the problem: objective function, constraints, design variables, optimizer and solver options

Optimizer interface		Aerostructural solver		Geometry modeler
pyOptSparse Common interface to various optimization software		*AeroStruct* Coupled solution methods and coupled derivative evaluation		*DVGeometry/GeoMACH* Defines and manipulates geometry, evaluates derivatives
SNOPT	Other optimizers	Flow solver *ADflow* Governing and adjoint equations	Structural solver *TACS* Governing and adjoint equations	

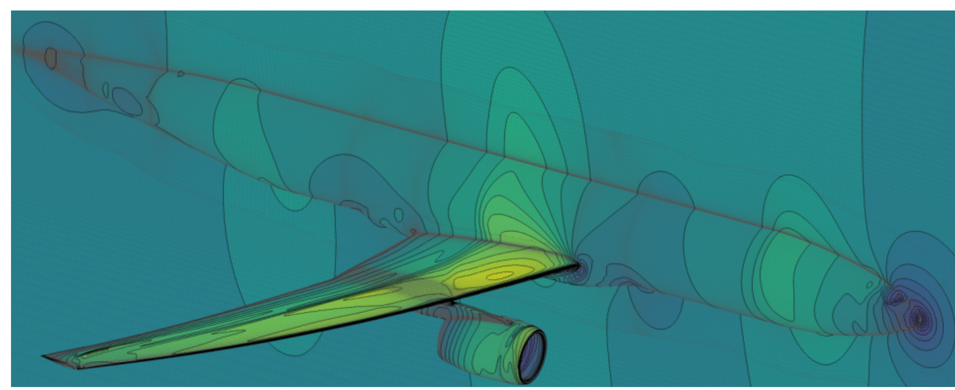
- Underlying solvers are parallelized and compiled
- All communication done through memory
- Easy-to-use Python scripting interface
- Only using aerodynamic design capacity for PADRI

ADFlow

- Automatic-Differentiation Flow Solver
- Second order finite volume RANS
- Standard SA turbulence model
- Point-matched multiblock and overset grids
- Multiple solvers: Runge Kutta (RK), DDADI, approximate Newton Krylov (ANK) and Newton Krylov (NK) algorithms
- DADI, ANK and NK used for optimization
- Extremely fast convergence for small design changes



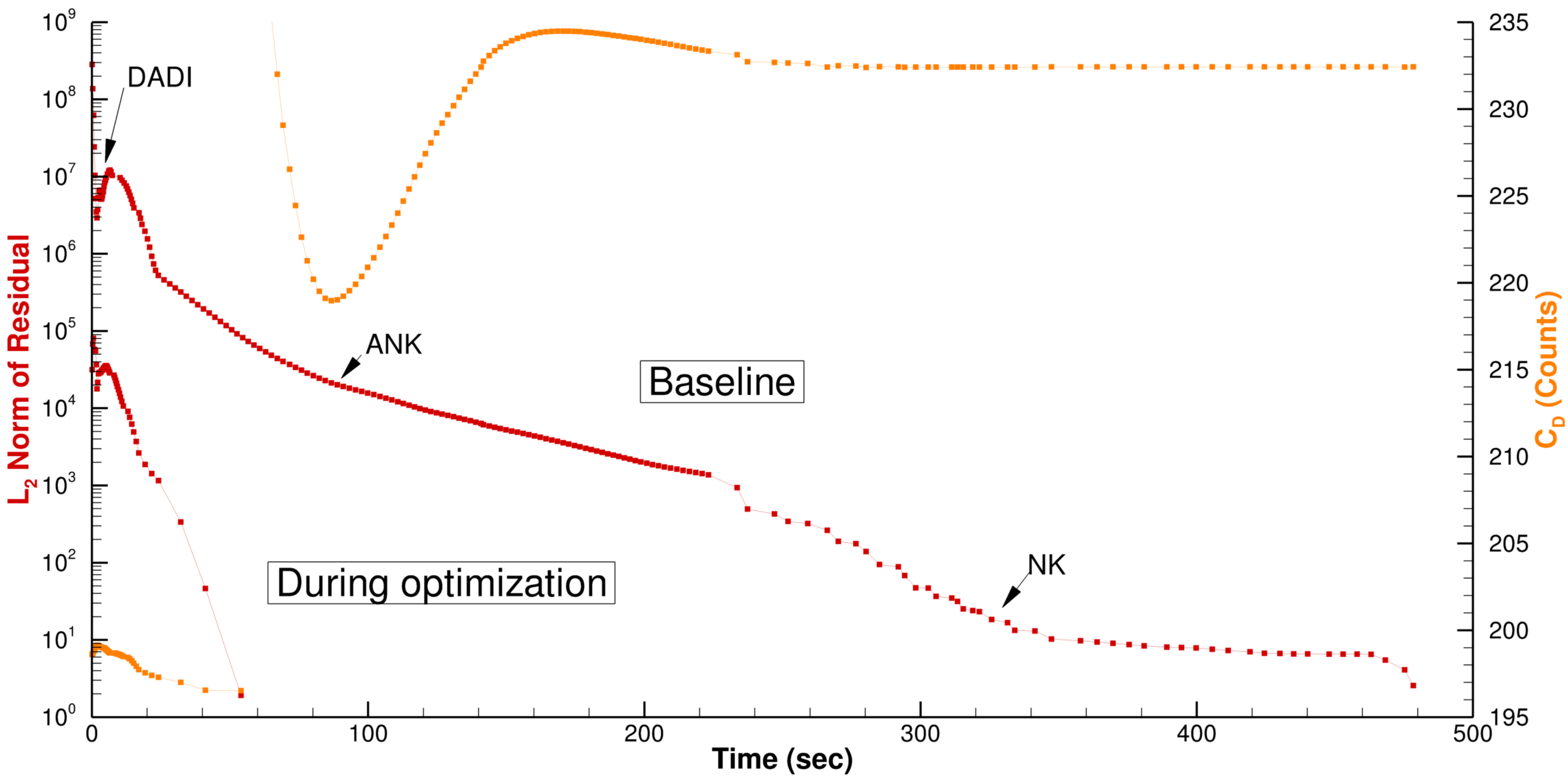
MIT D8 Double Bubble



Common Research Model (DPW6)

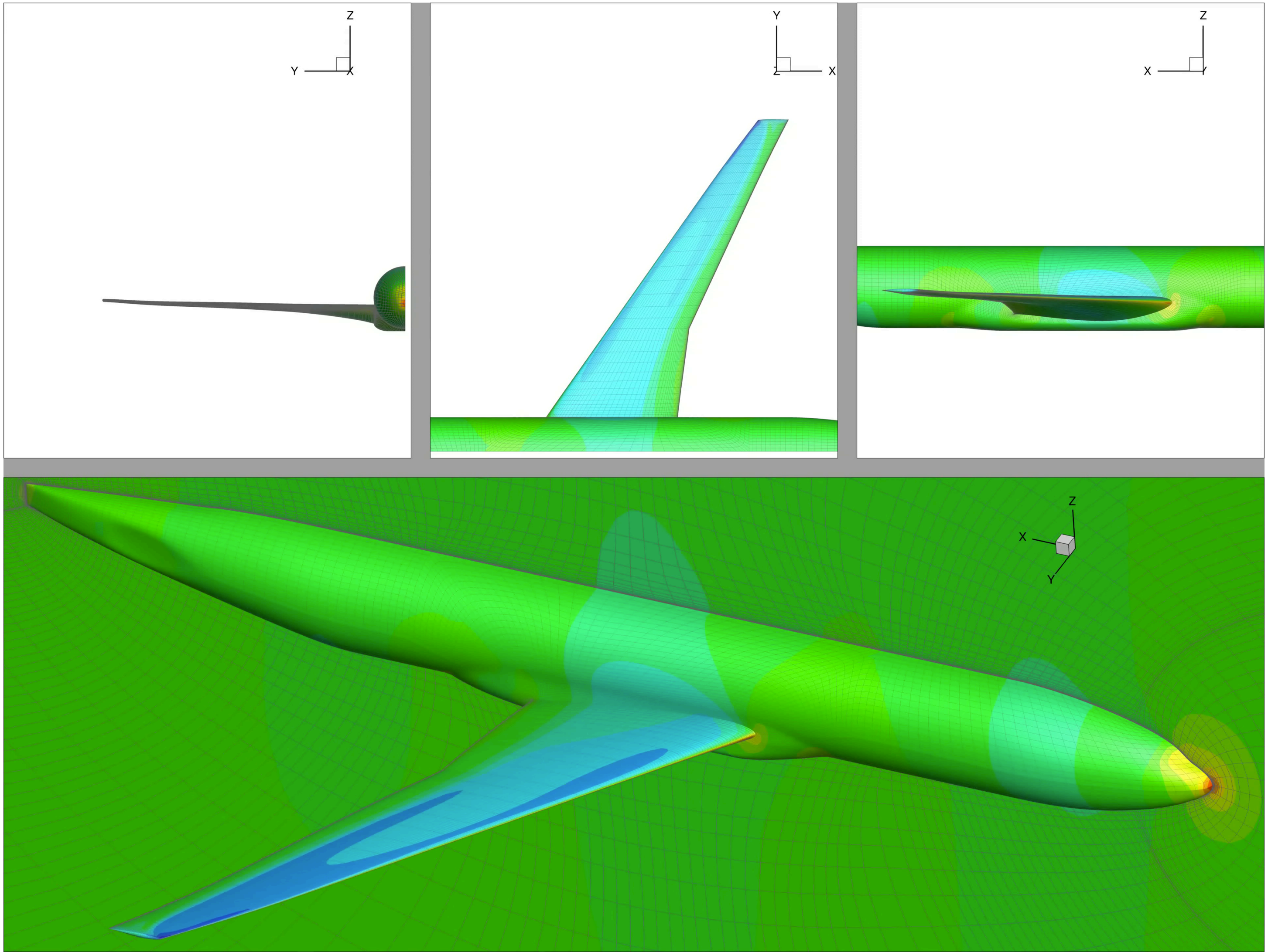
ADFlow Solver Convergence

- Combination of three algorithms: Diagonalized Alternating Direction Implicit (DADI), Approximate Newton-Krylov (ANK) and Newton Krylov (NK)
- Newton-Krylov fully couples flow and turbulence variables



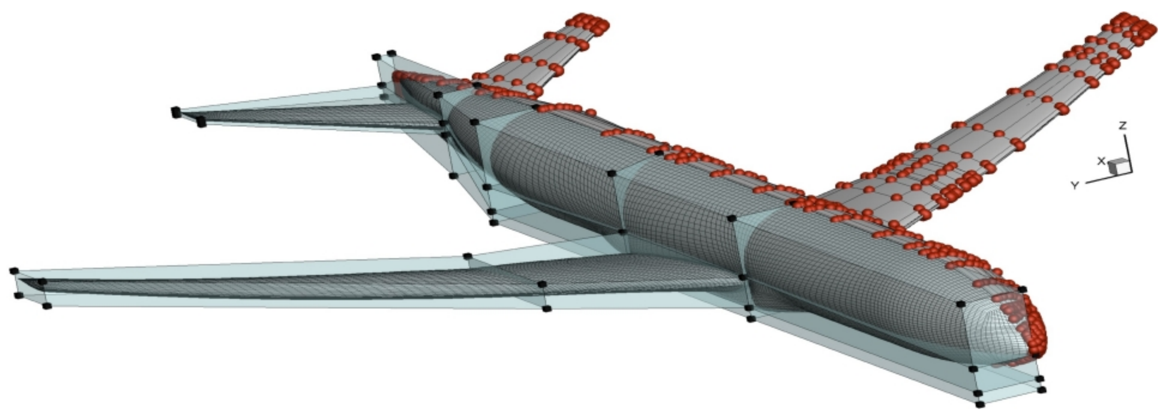
Mesh Deformation

- Inverse-distance weighting method
- Parallel, fast and highly robust for large deformations



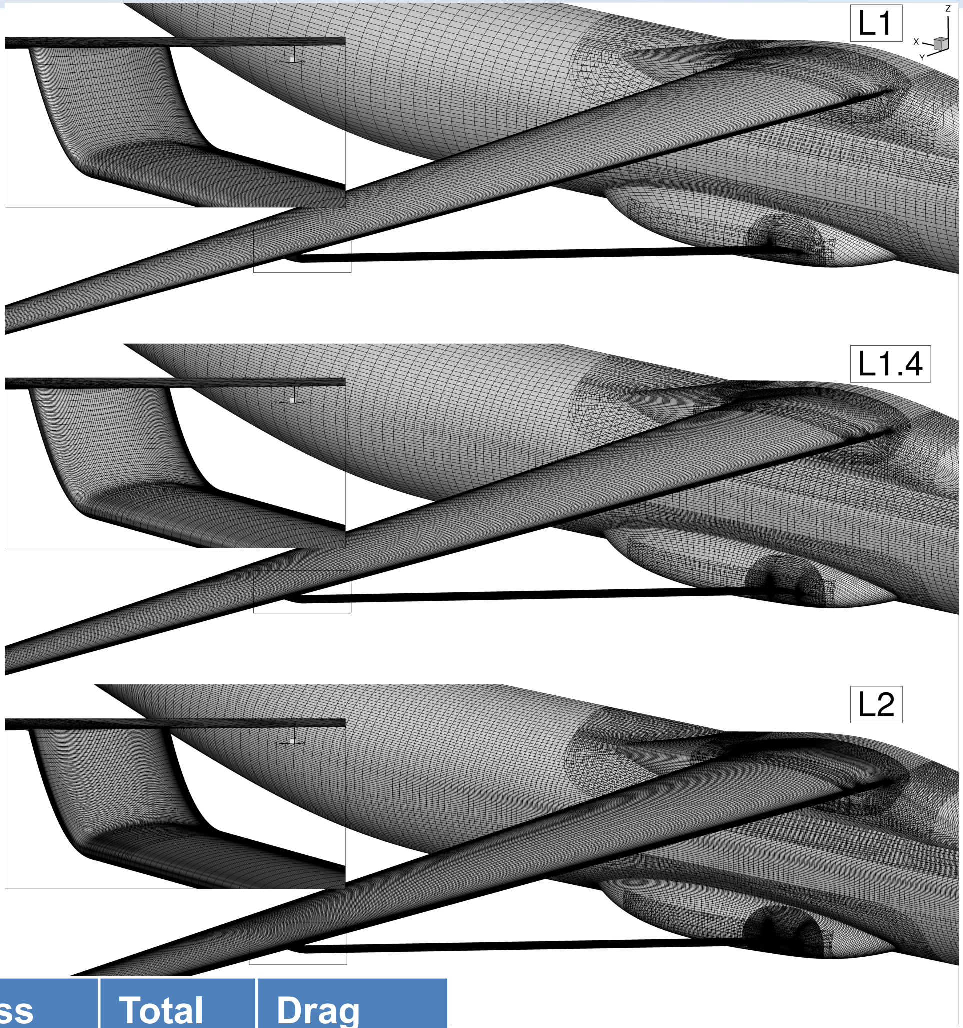
Geometry Manipulation

- Free-form deformation (FFD) volume approach
- Parametrize the change in geometry
- Embed discrete geometry into trivariate B-spline volumes
- Point-inversion algorithm to find u-v-w coordinates
- Control point motion smoothly controls the underlying geometry
- Sub-FFD approach for localized control



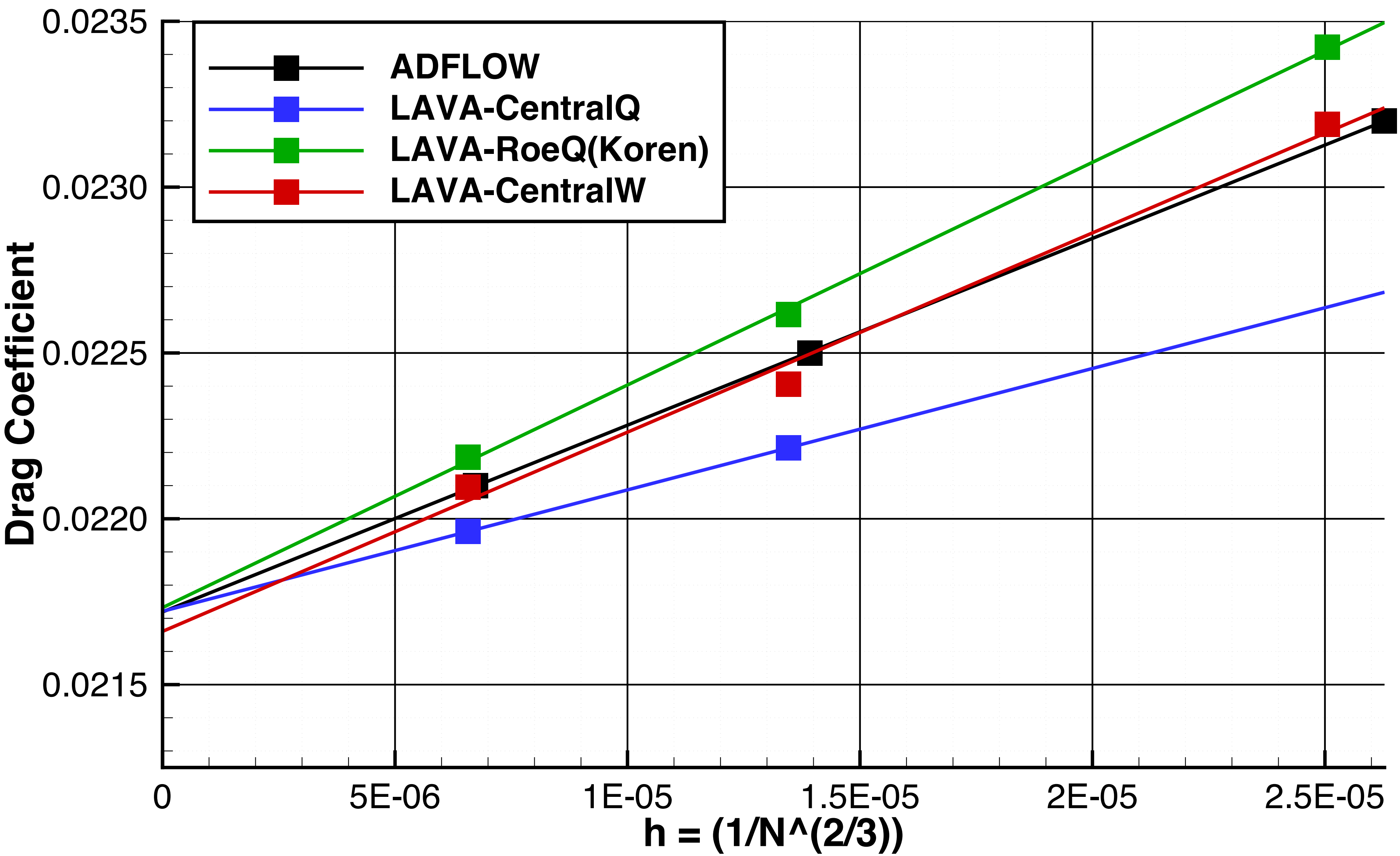
Overset Meshes

- Surface patches generated with Pointwise
- Chimera Grid Tools (CGT) for volumetric extrusion
- Hyperbolic mesh extrusion
- Consistent refinement between levels

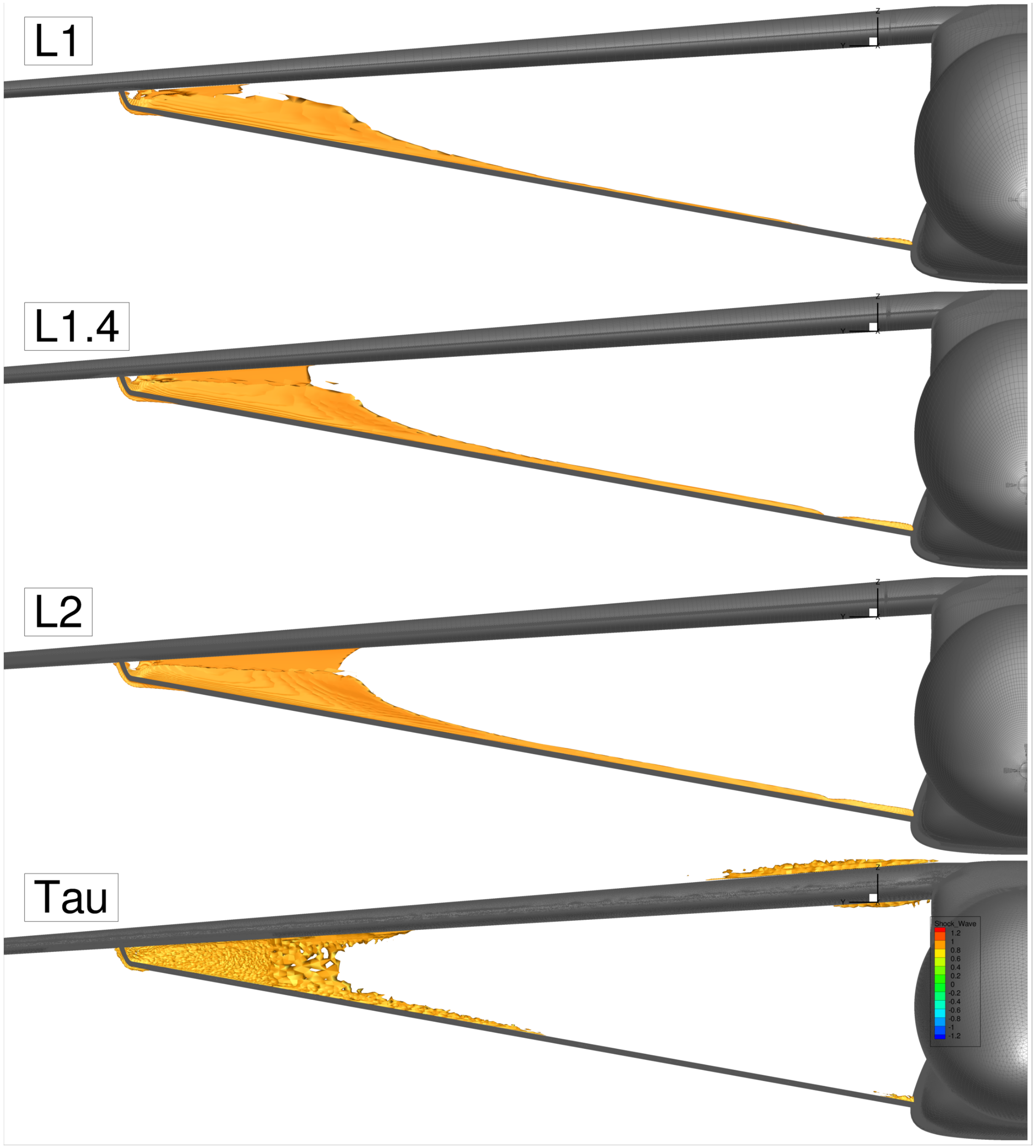


Mesh	# Wing Chordwise	# Wing Spanwise	# Truss Chordwise	# Truss Spanwise	Total Cells	Drag (counts)
L1	64	202	96	110	7.4 M	232.42
L1.4	88	282	134	154	19.2 M	224.61
L2	126	404	192	220	57.3 M	220.87

Baseline Configuration Grid Convergence



Baseline Solutions (Shock Sensor)



Optimization Problem Description

- Single point drag minimization (CL=0.417)
- Design Variables: FFD Shape position + angle of attack
- Flight condition: M=0.72, altitude=30,000 ft, alpha=1.0

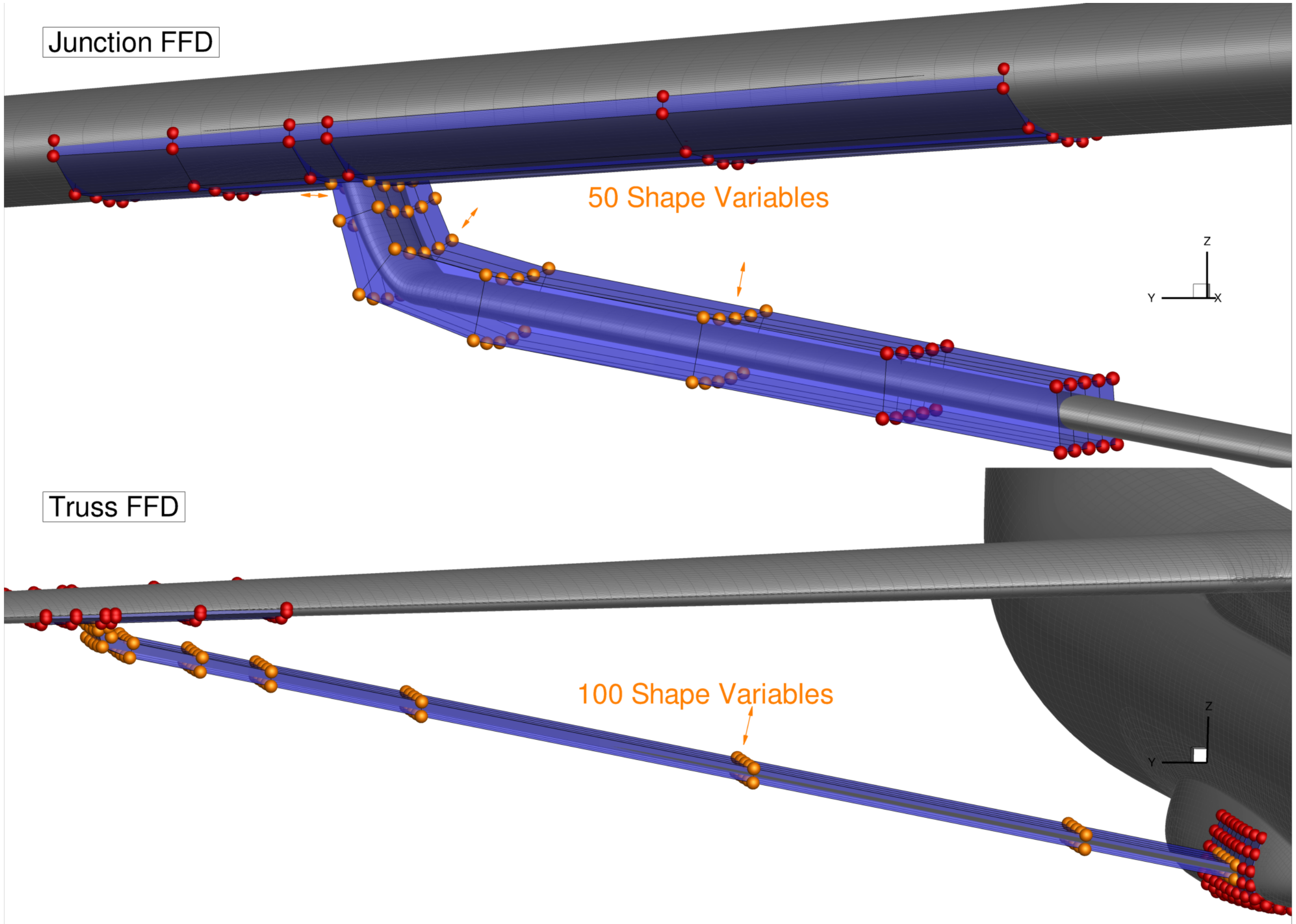
- Case 1
- Nominal design problem

Case 2
- Nominal design problem + fixed trailing edge

Case 3
- Full truss redesign

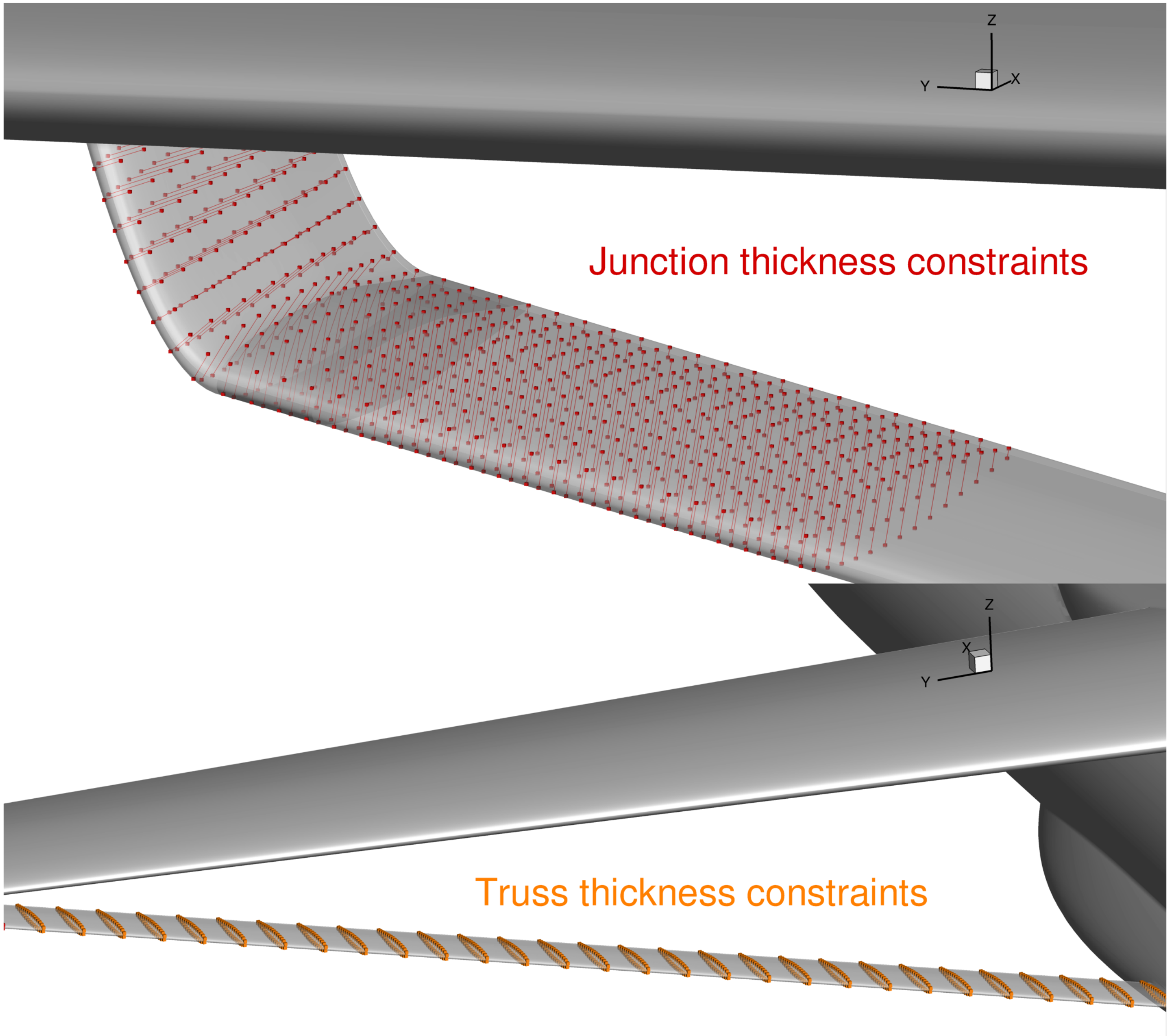
Optimization Design Variables

- Only truss is modified
- Follows workshop guidelines for design region (Case 1 and 2)
- Orange control point sphere are modified



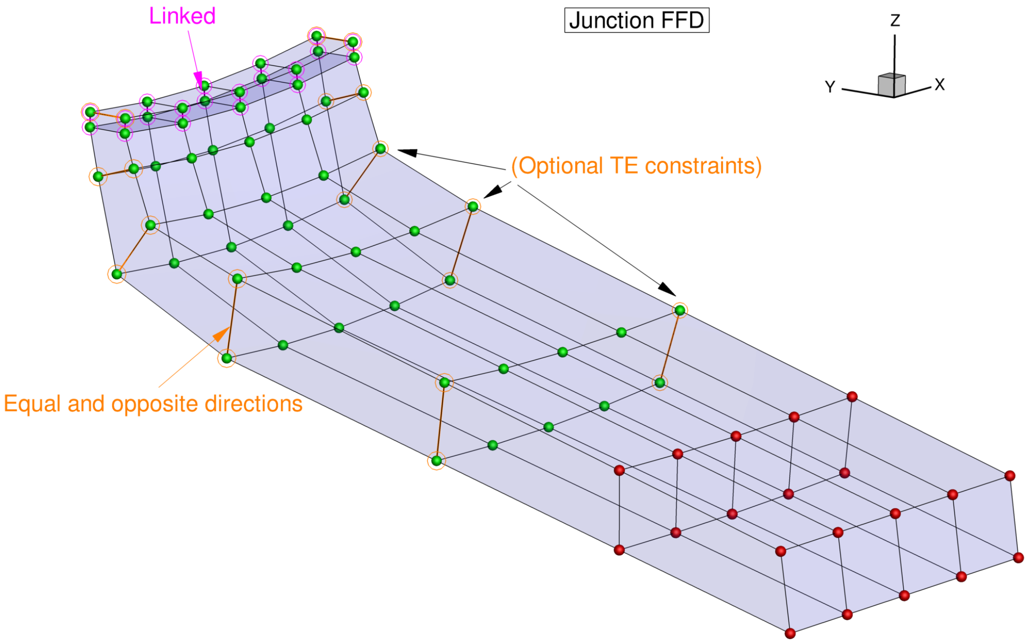
Optimization Constraints

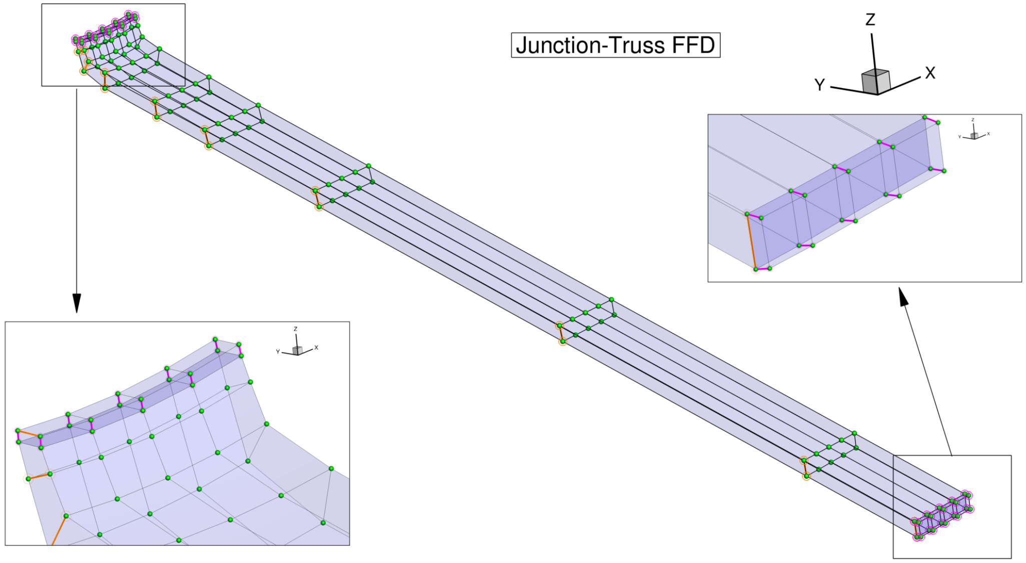
- Explicit "toothpick" thickness constraints



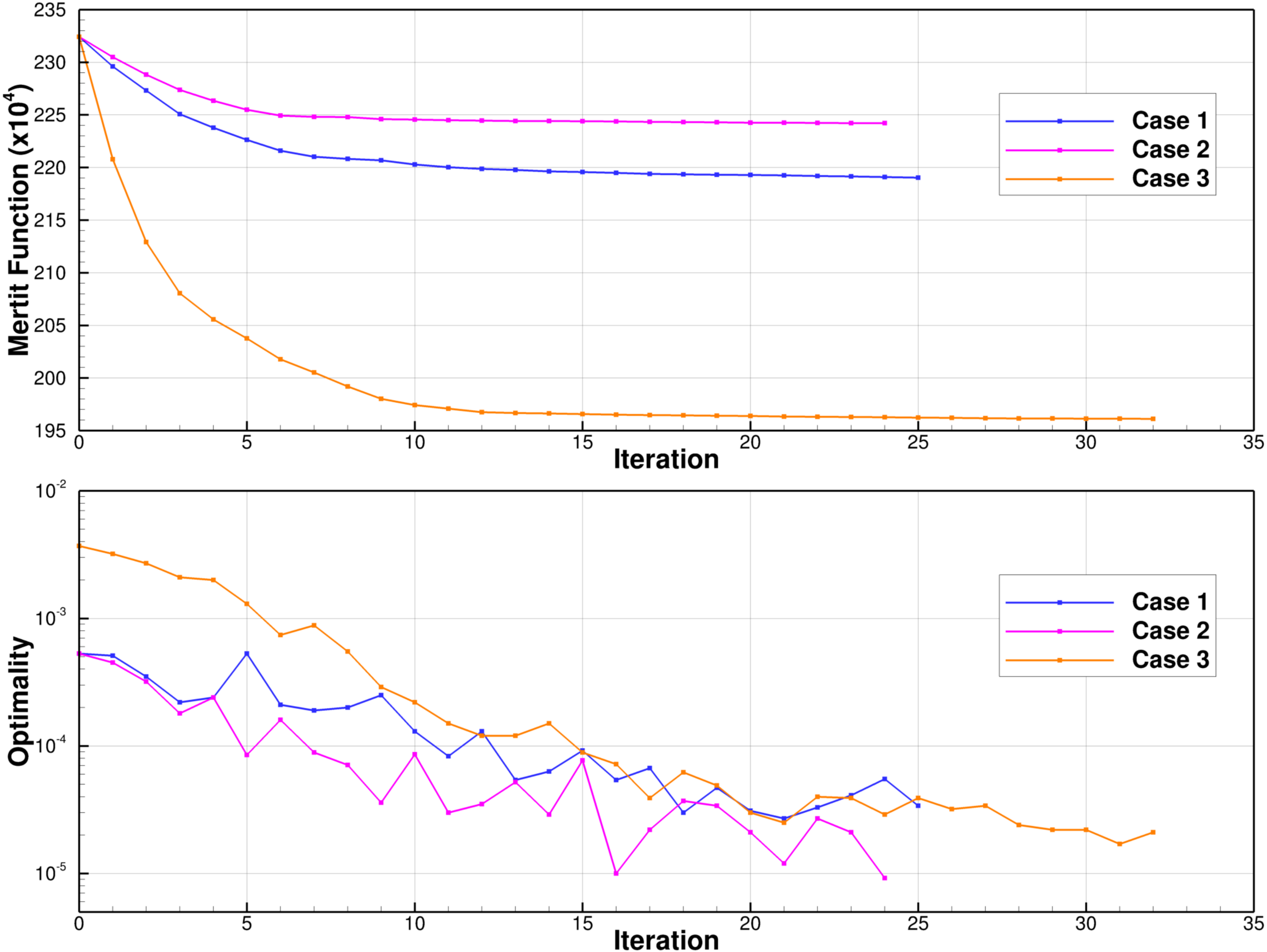
Optimization Constraints

- Linear constraints enforce fixed leading and (optionally) trailing edge
- These constraints are enforced exactly by the optimizer



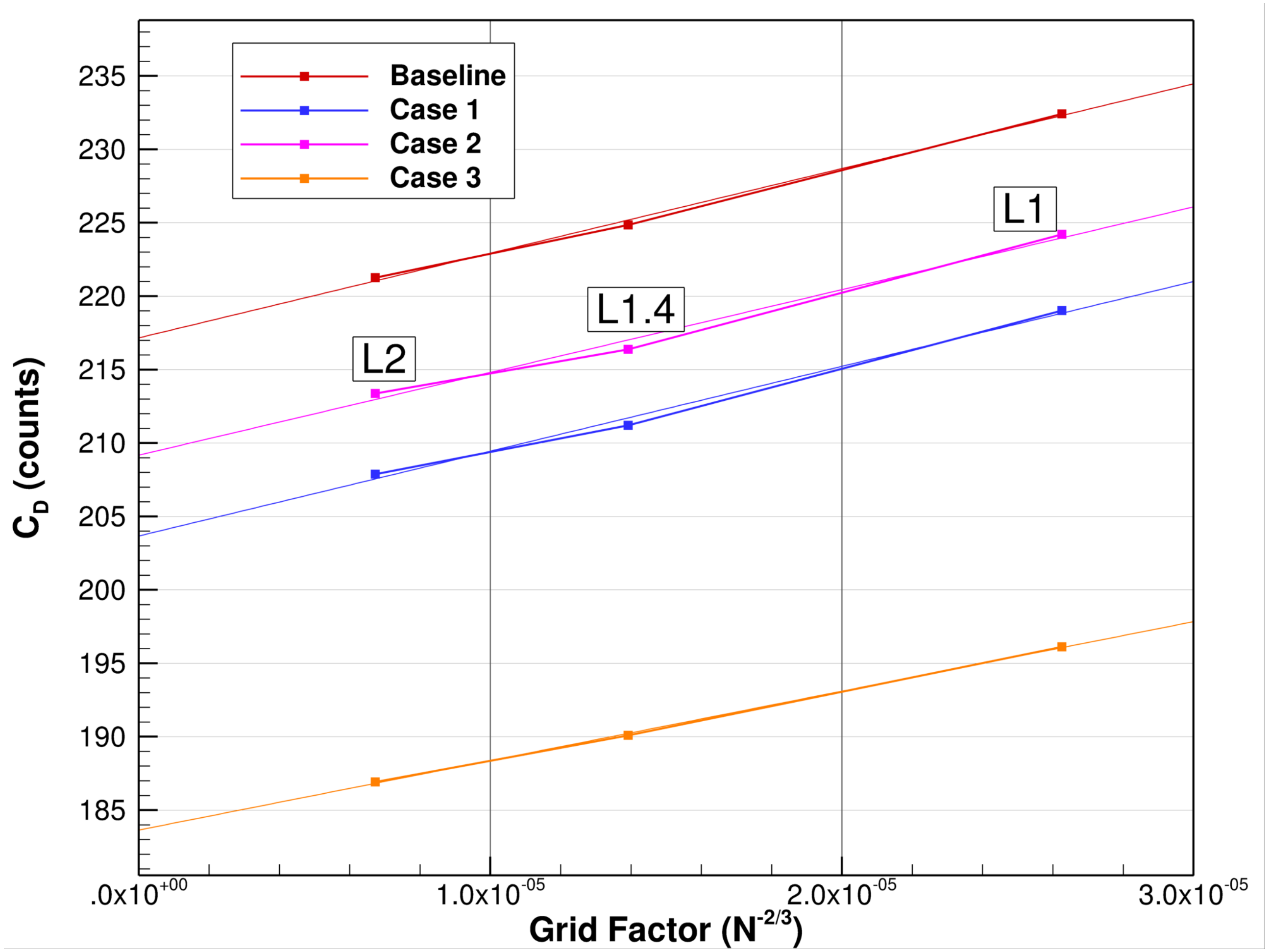


Optimization Convergence History



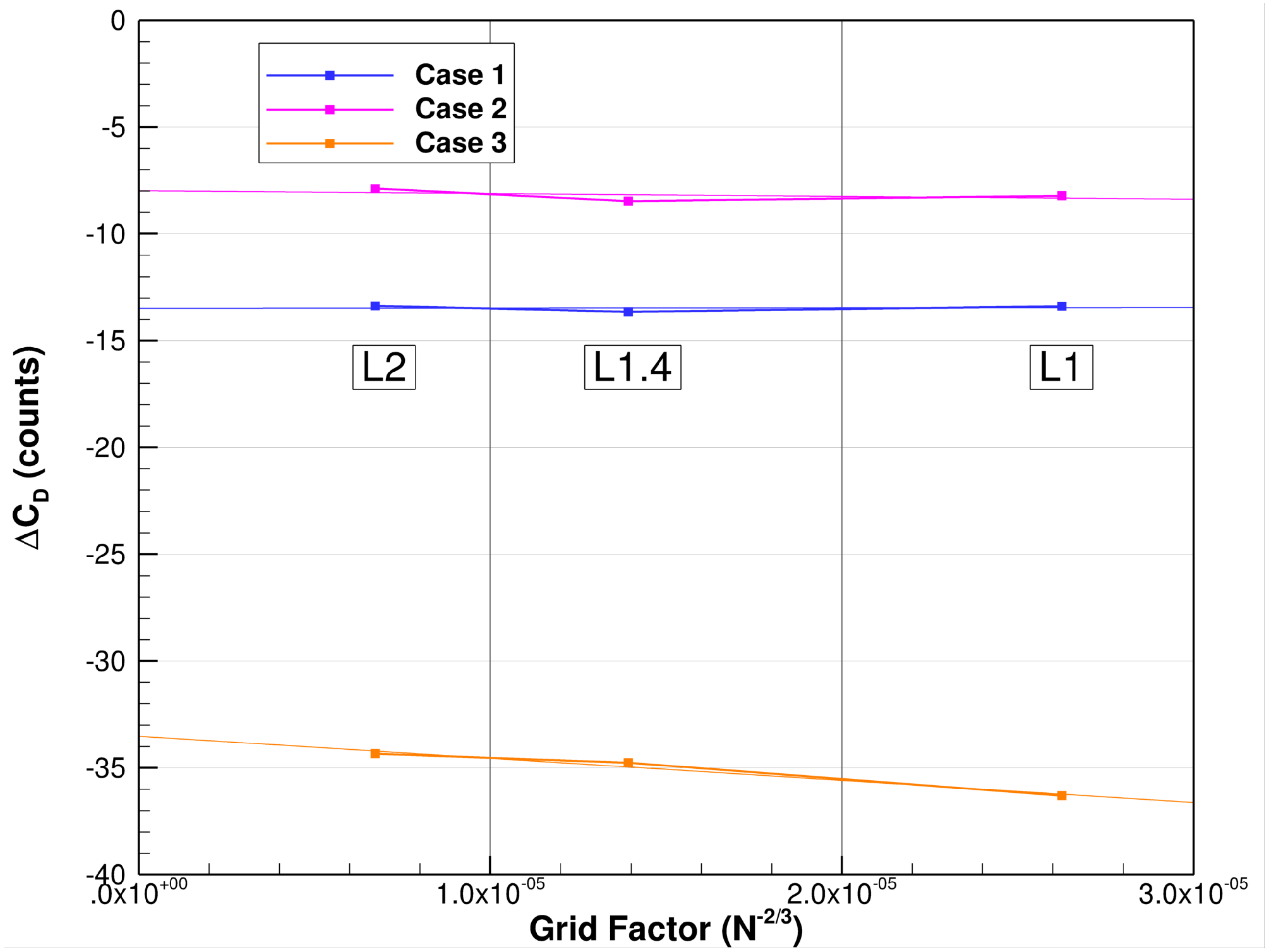
Grid Convergence Study

- Optimized L1 shape analyzed using finer meshes

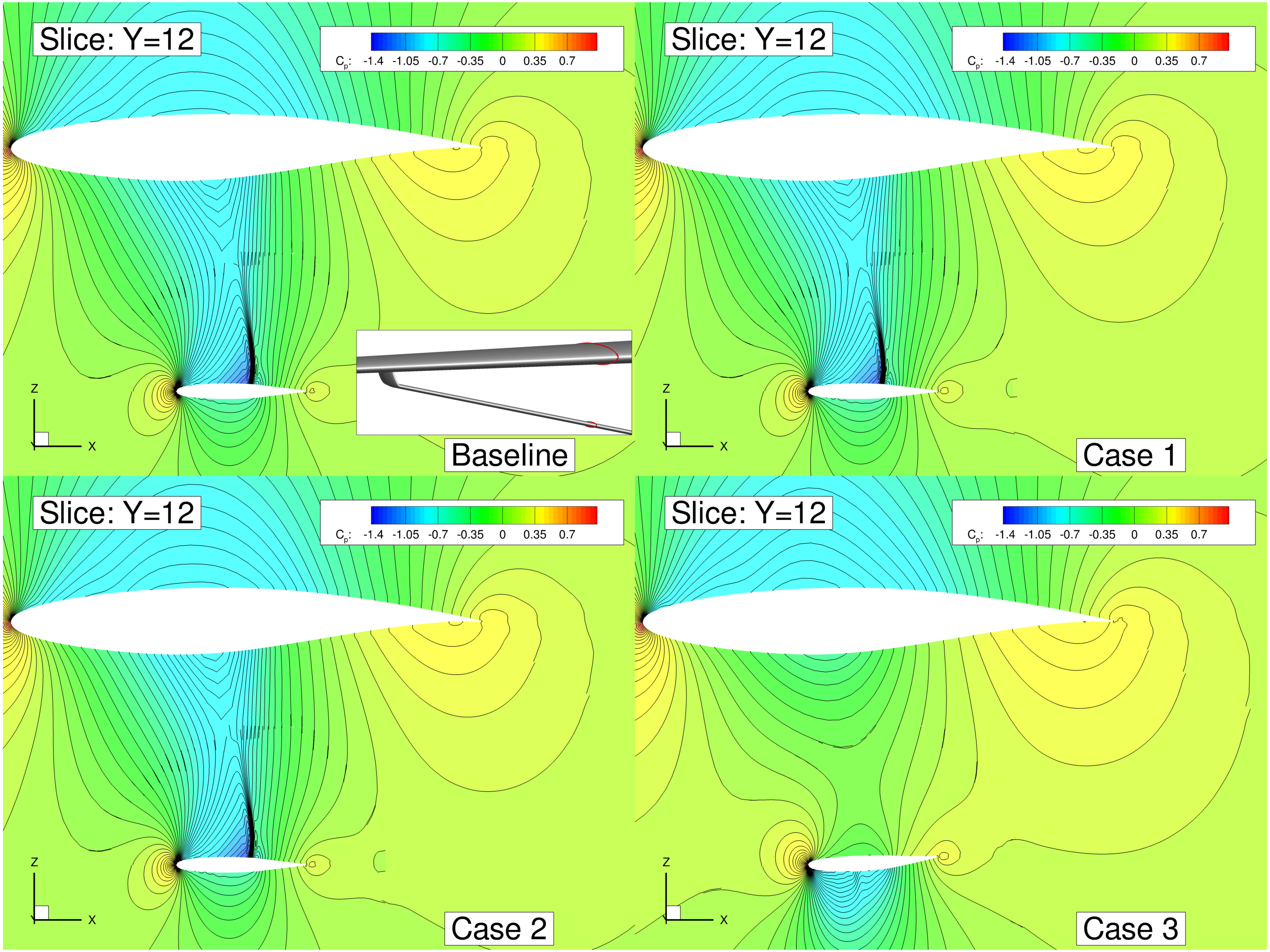


Grid Convergence Study

- Nearly constant drag deltas
- L1 mesh capturing the critical flow features



2D Slices of Junction Region



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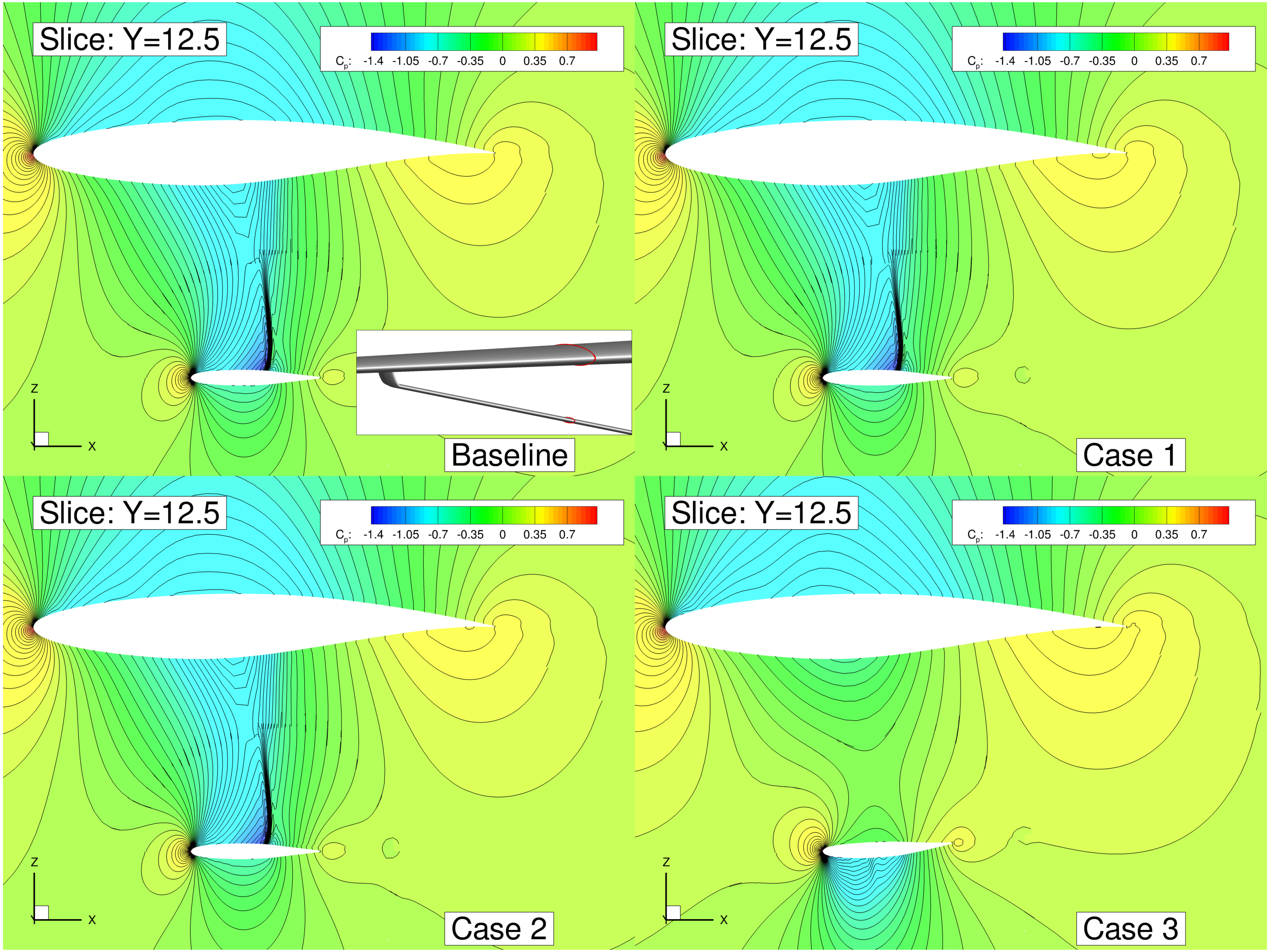

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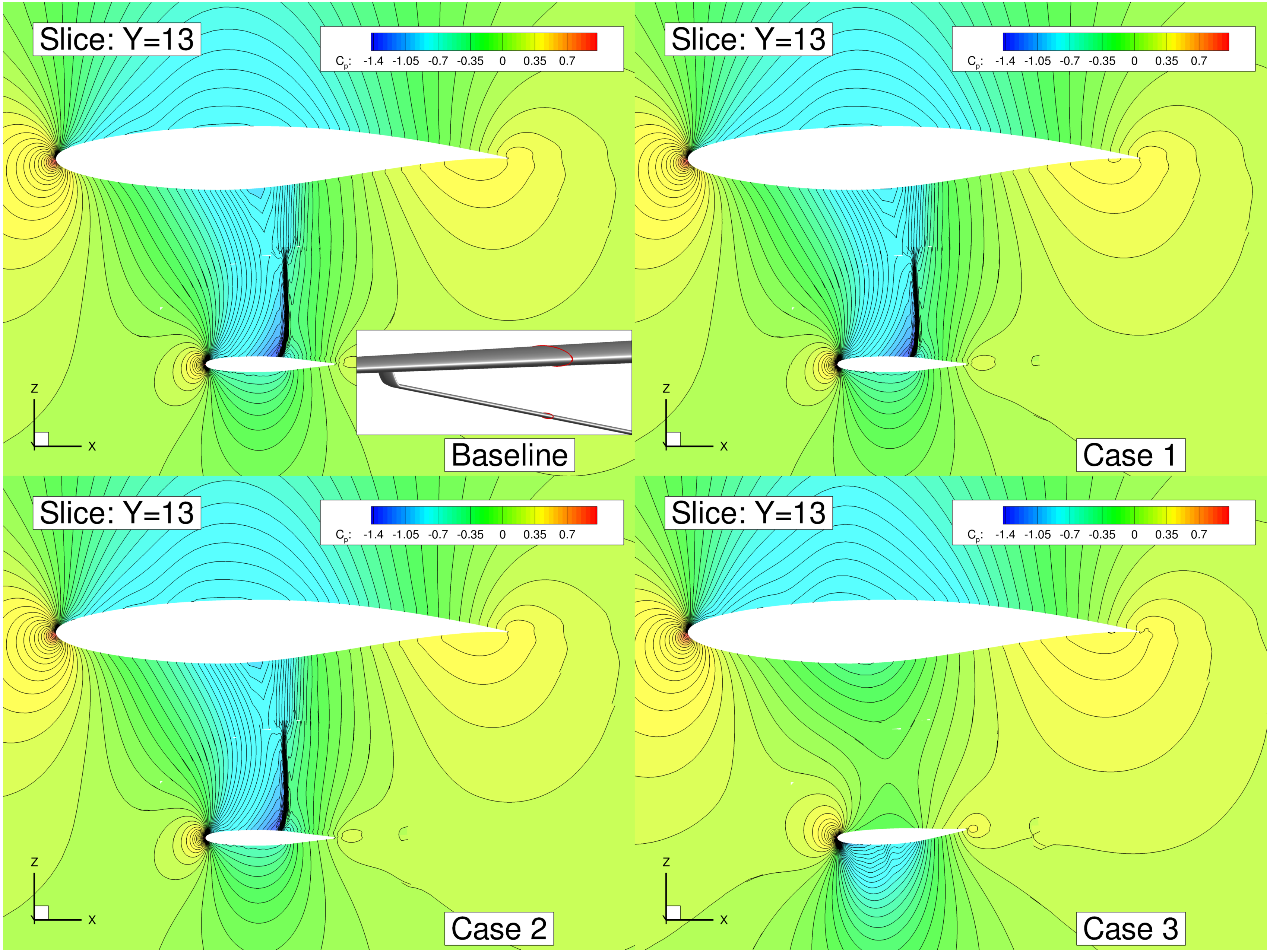

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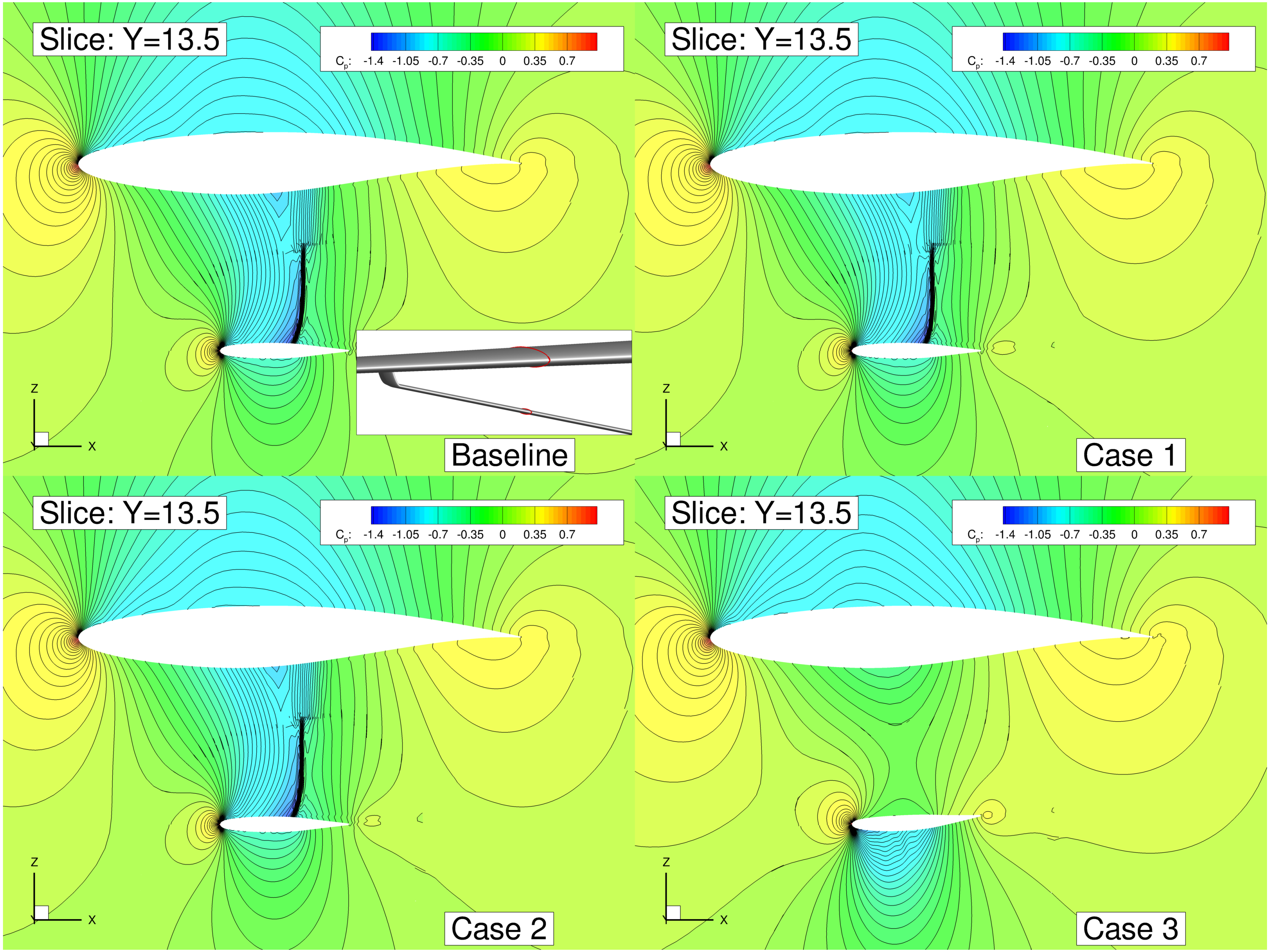

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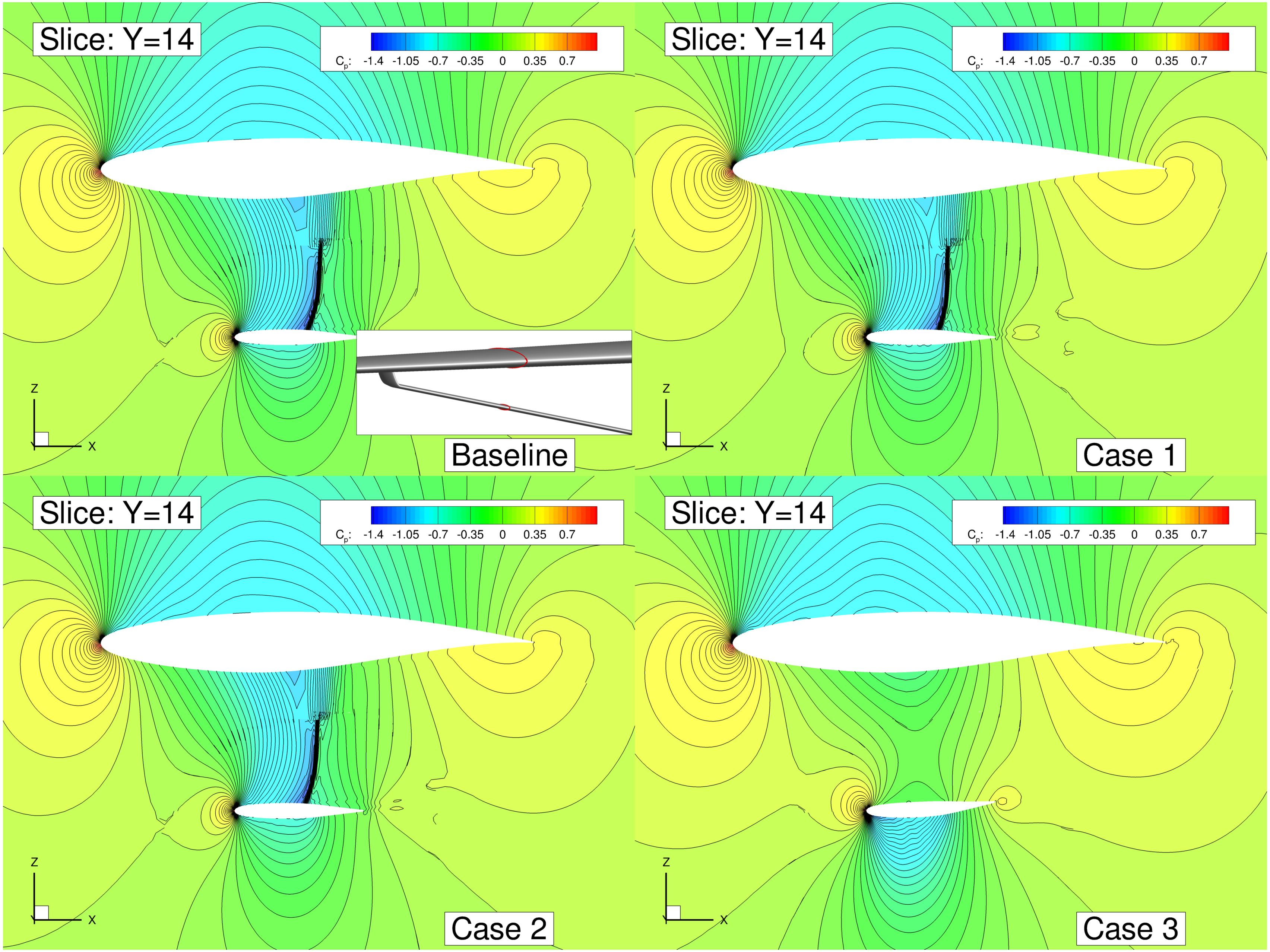

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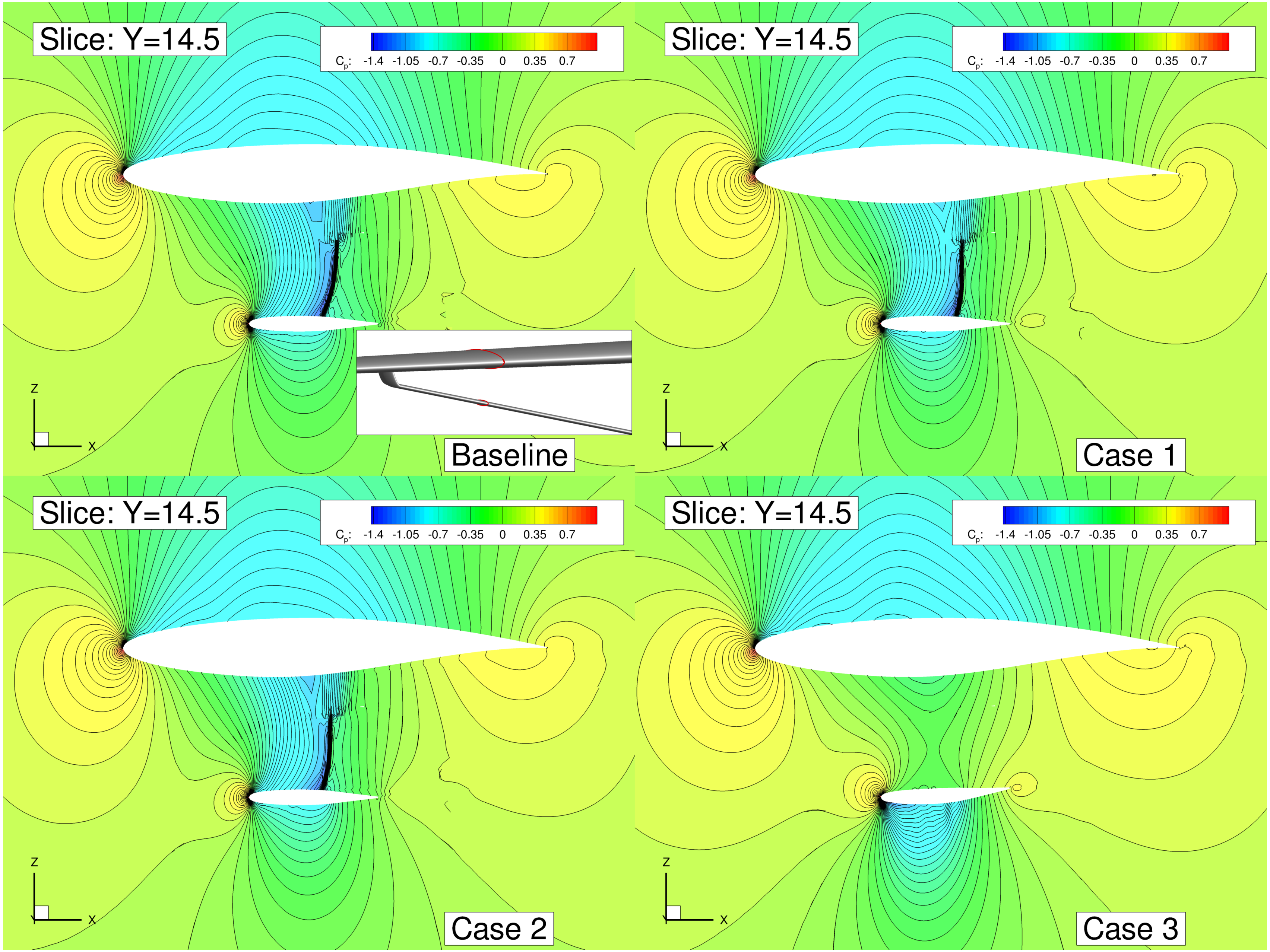

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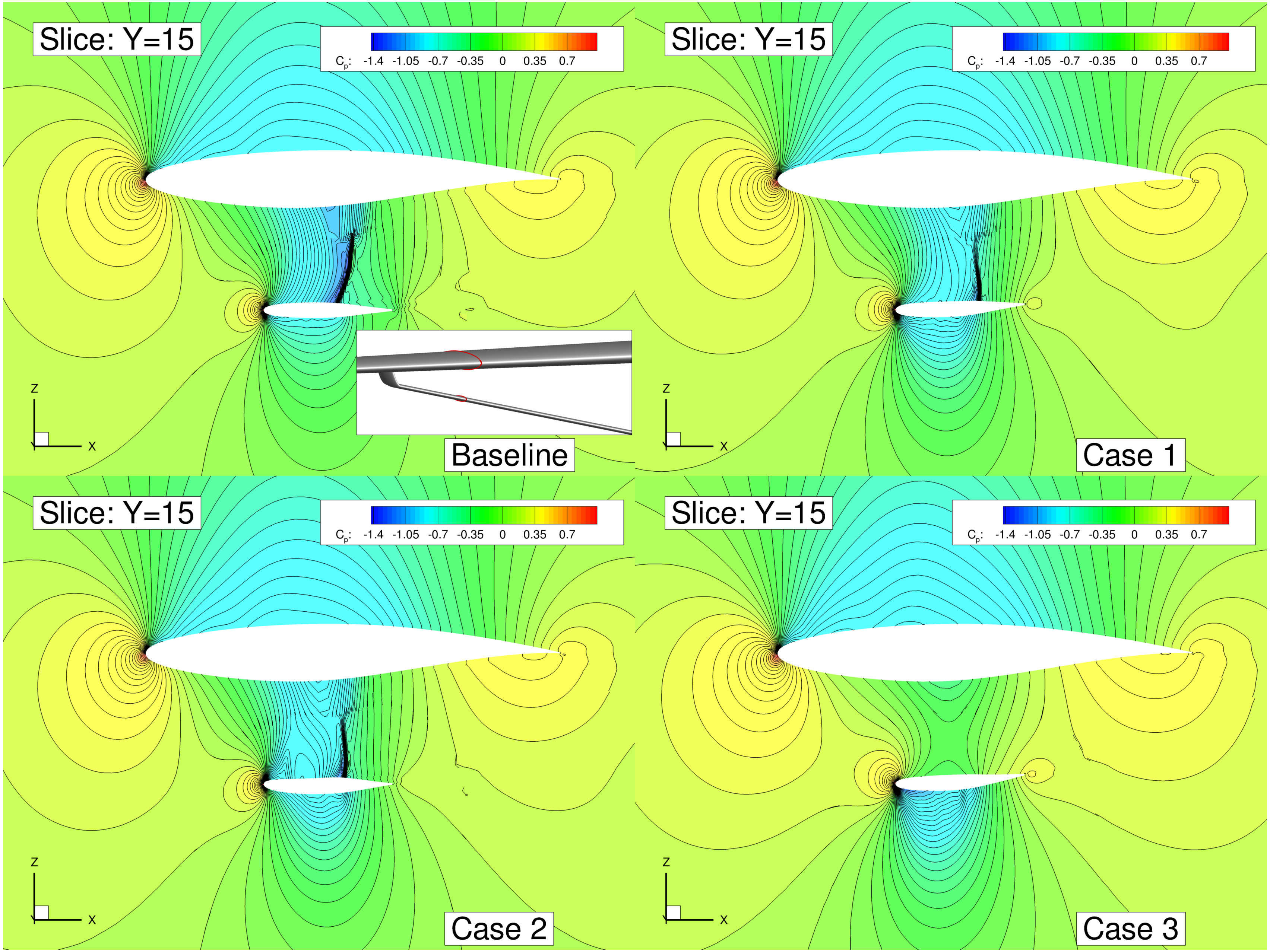

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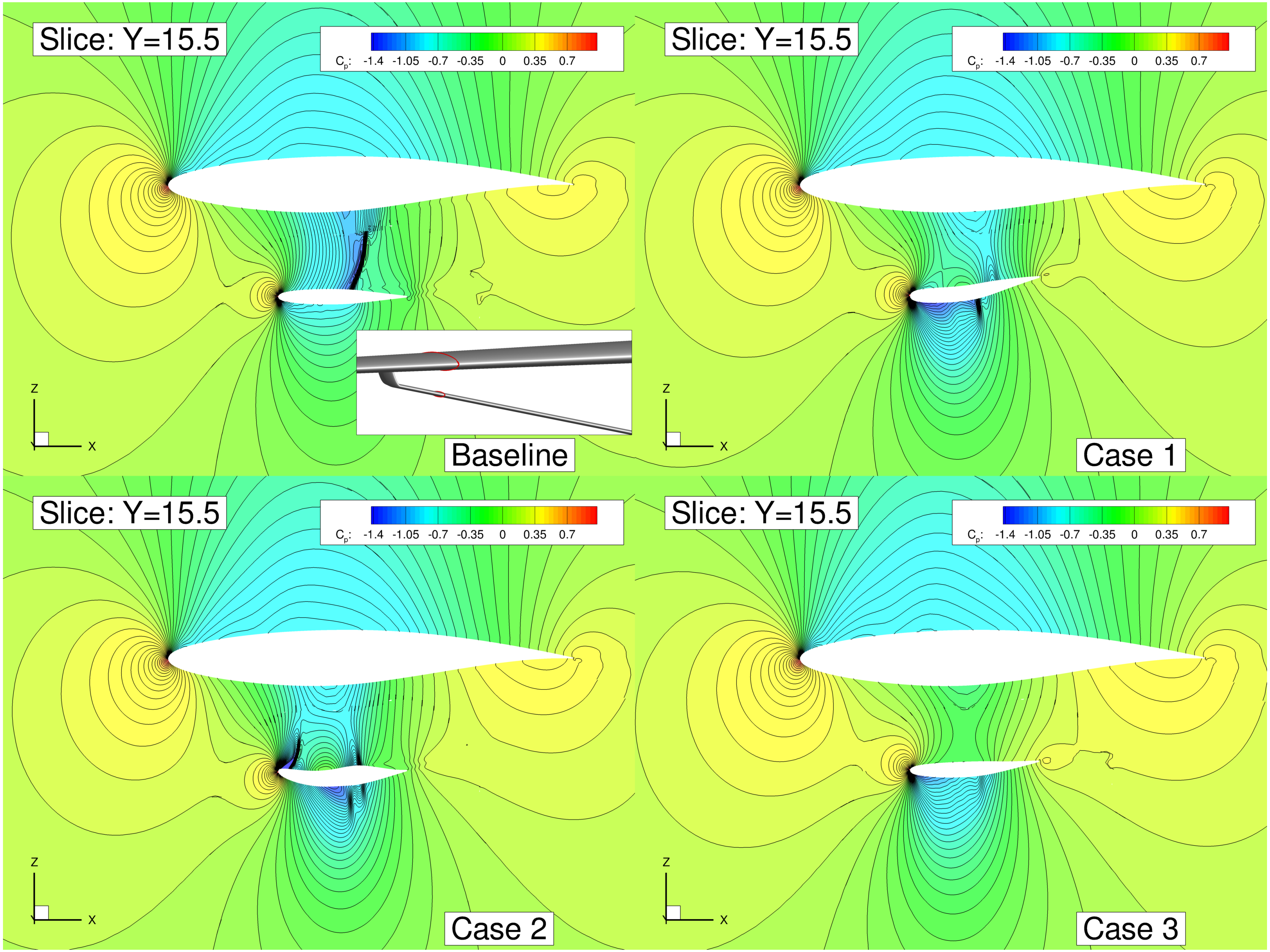

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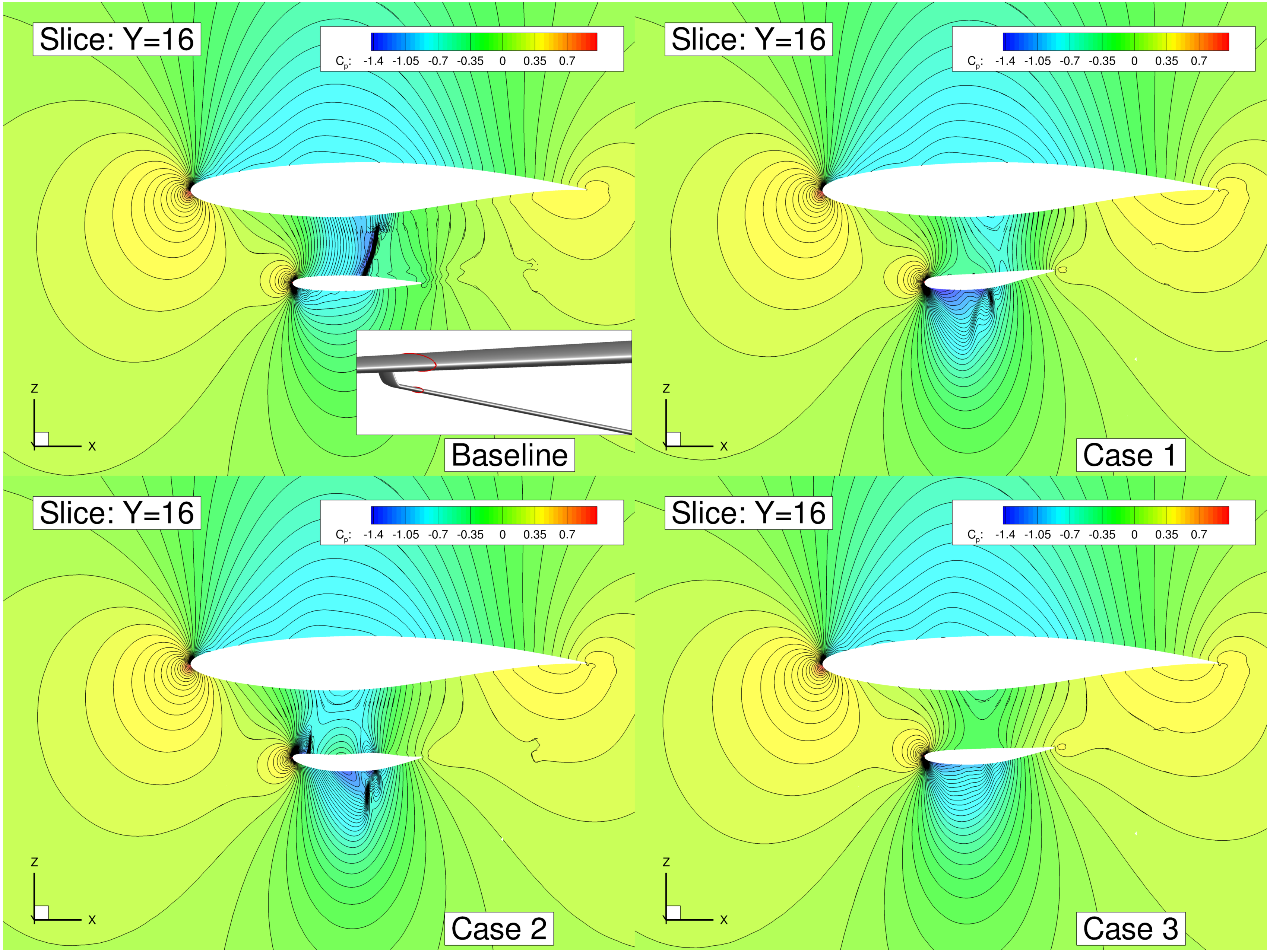

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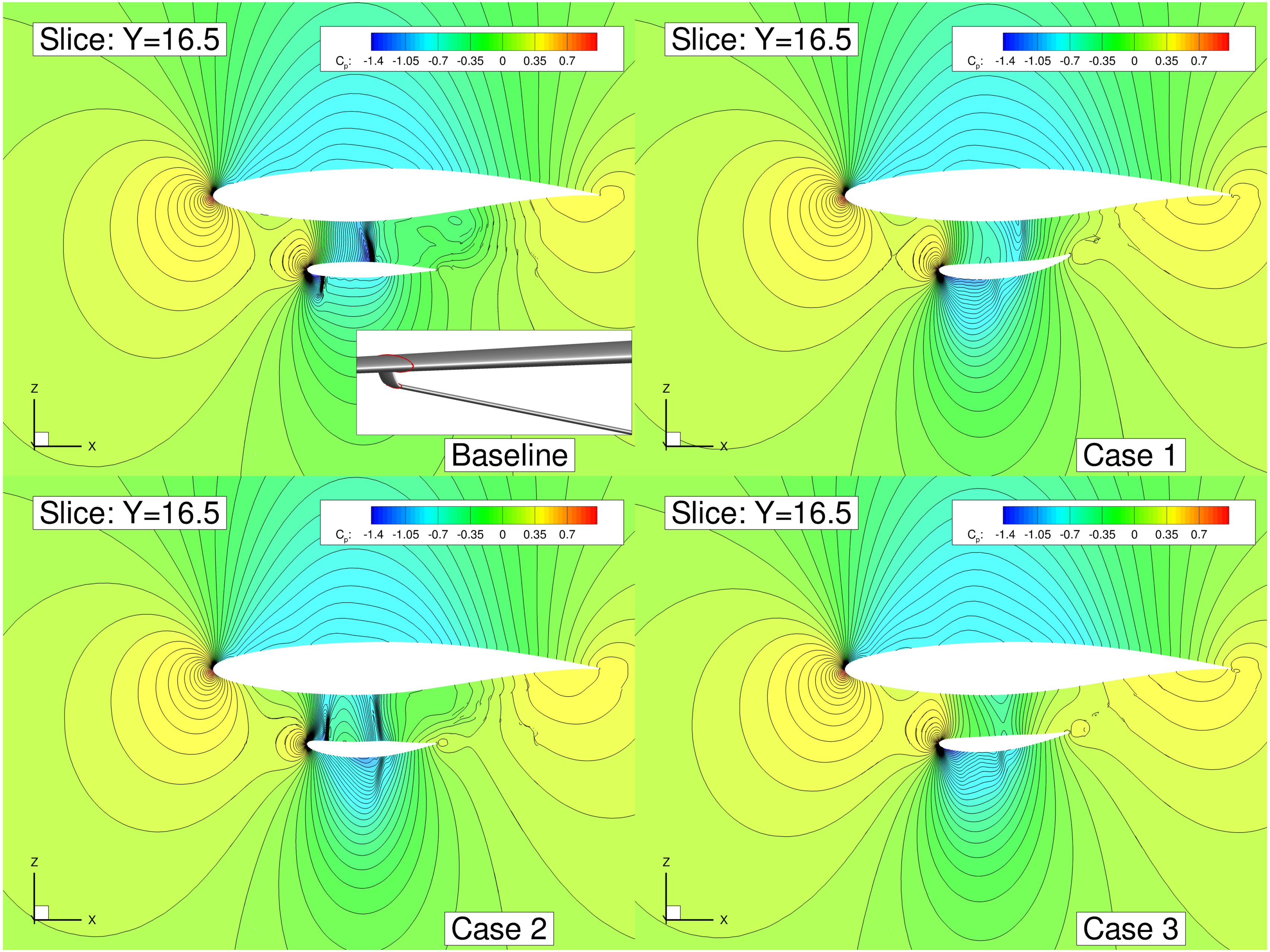

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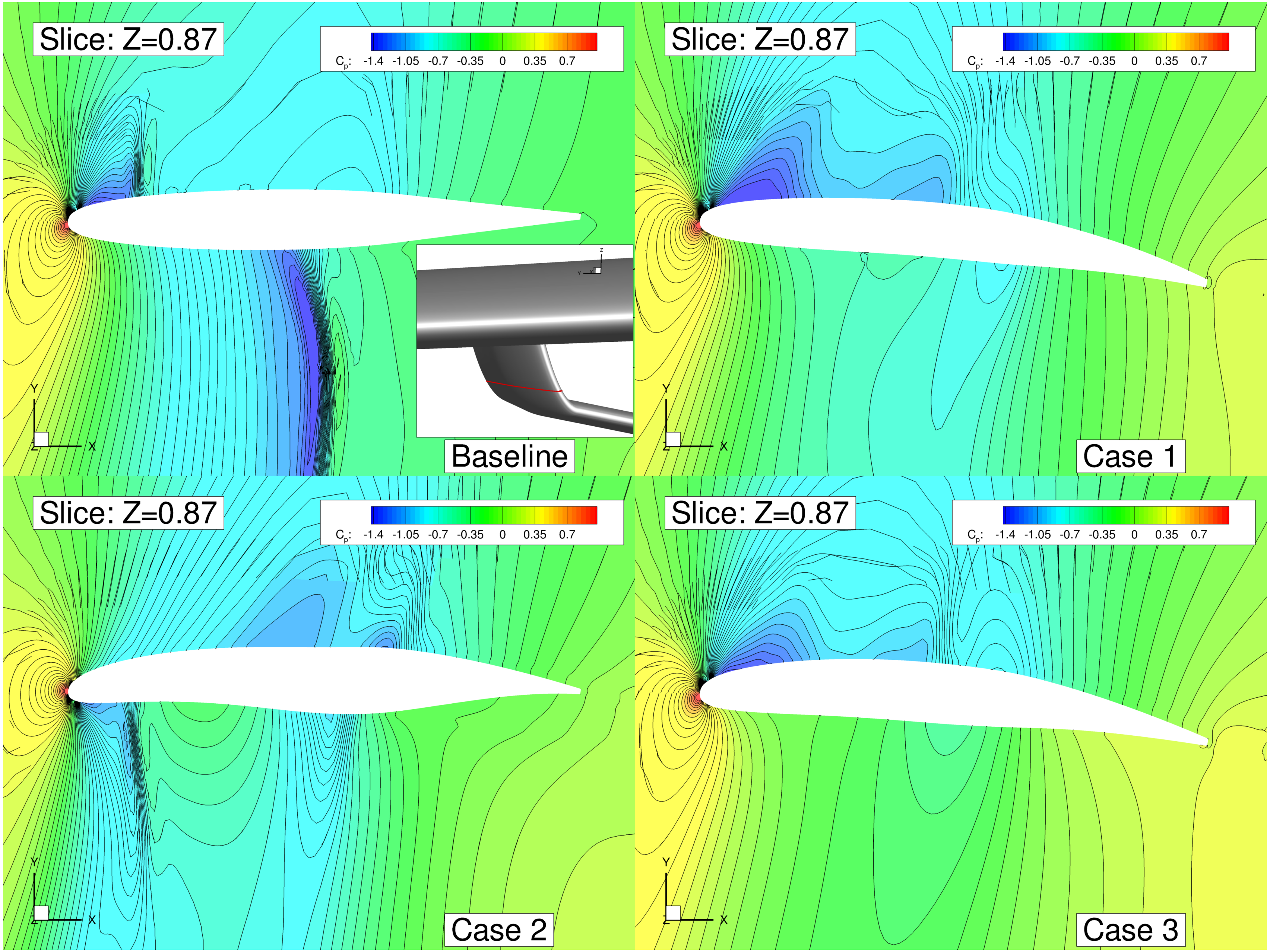

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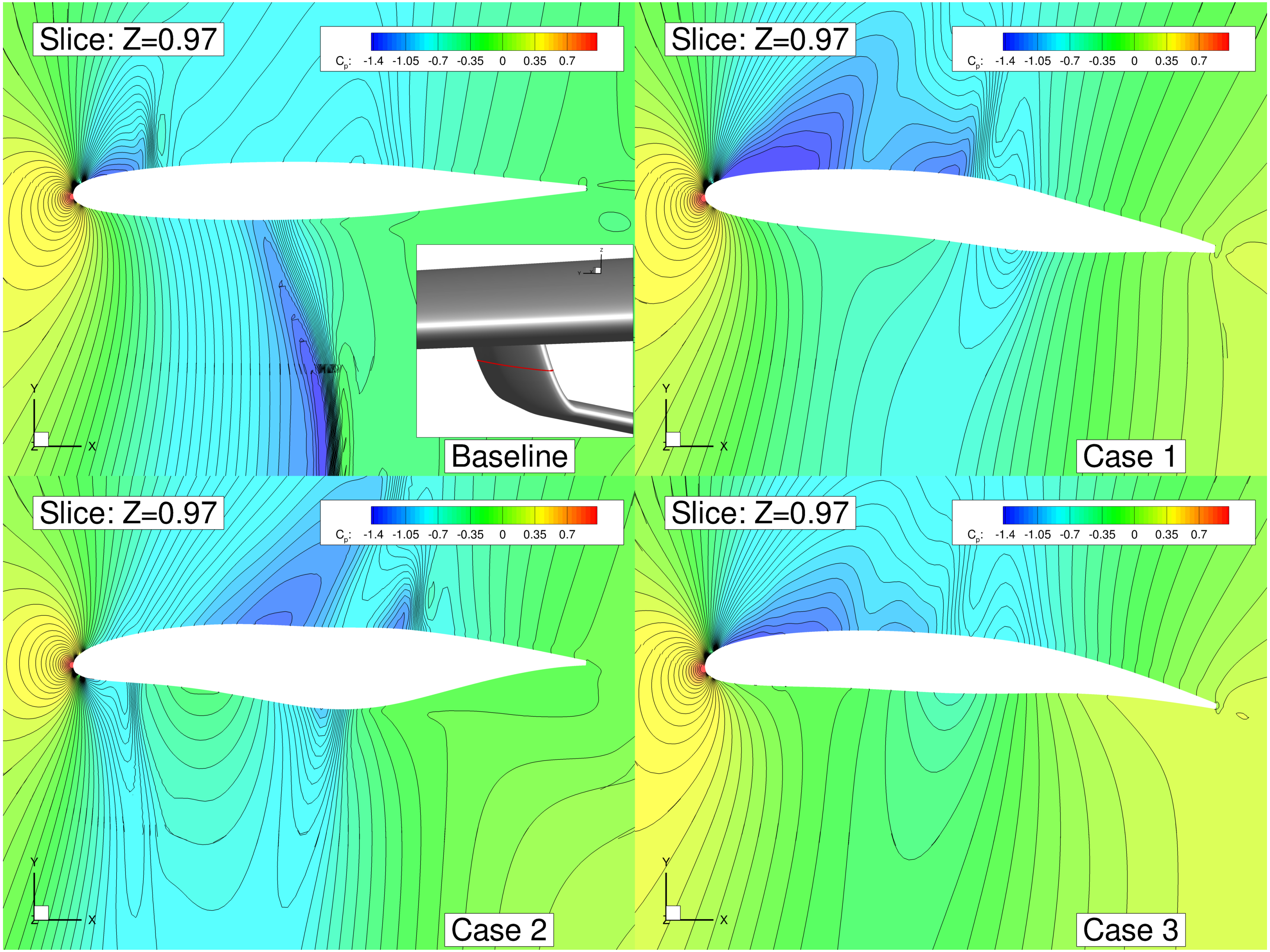

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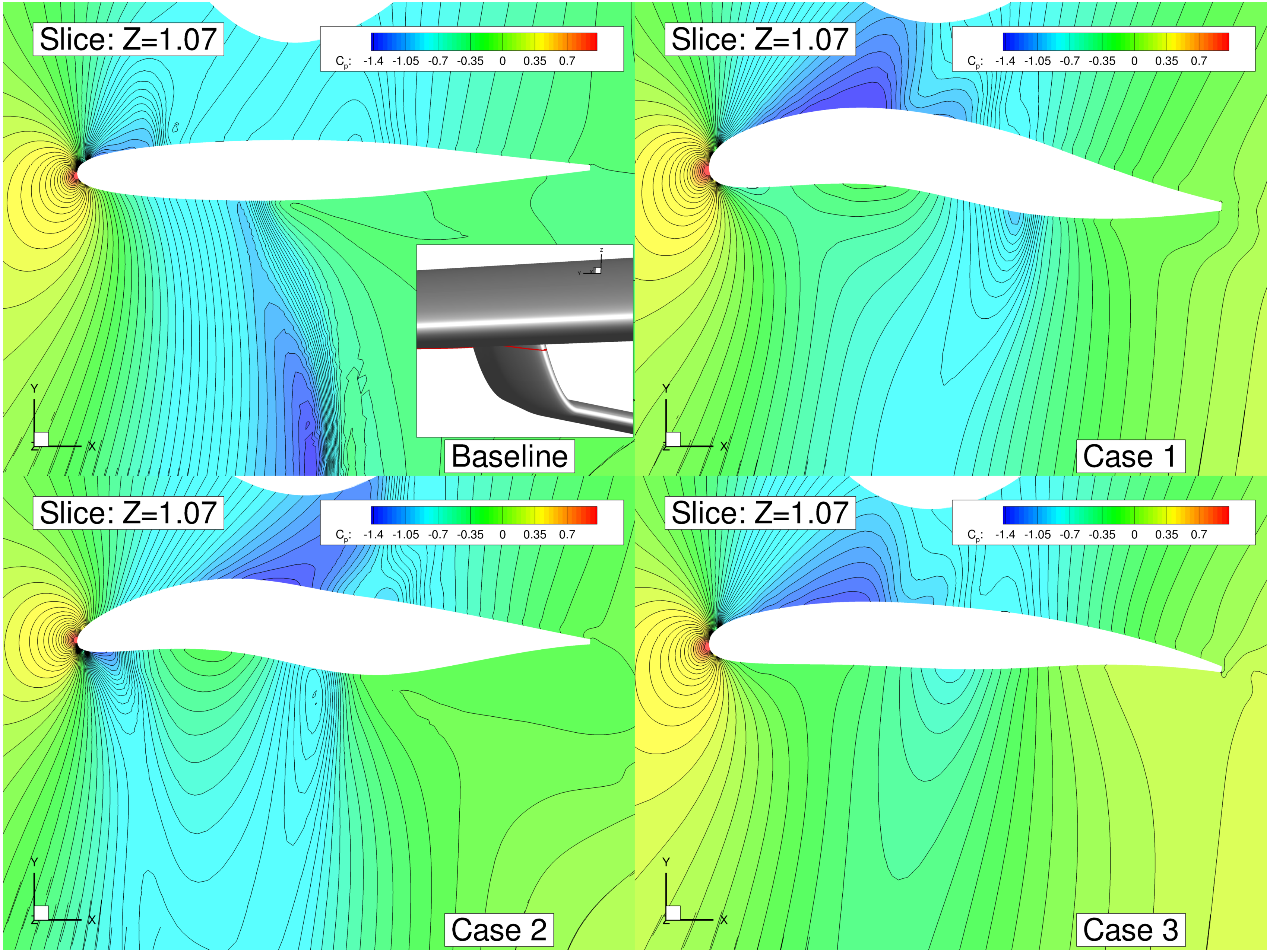

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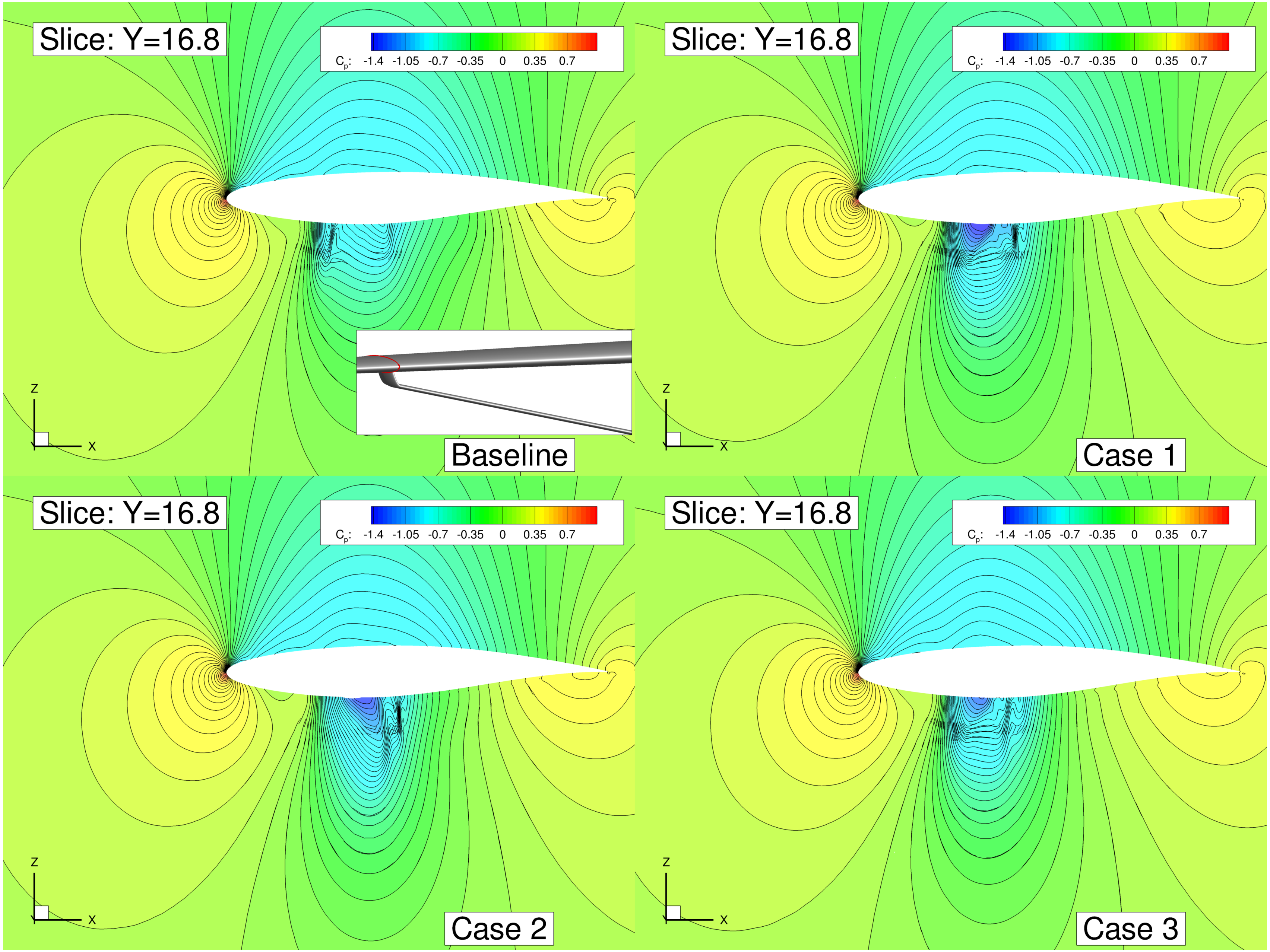

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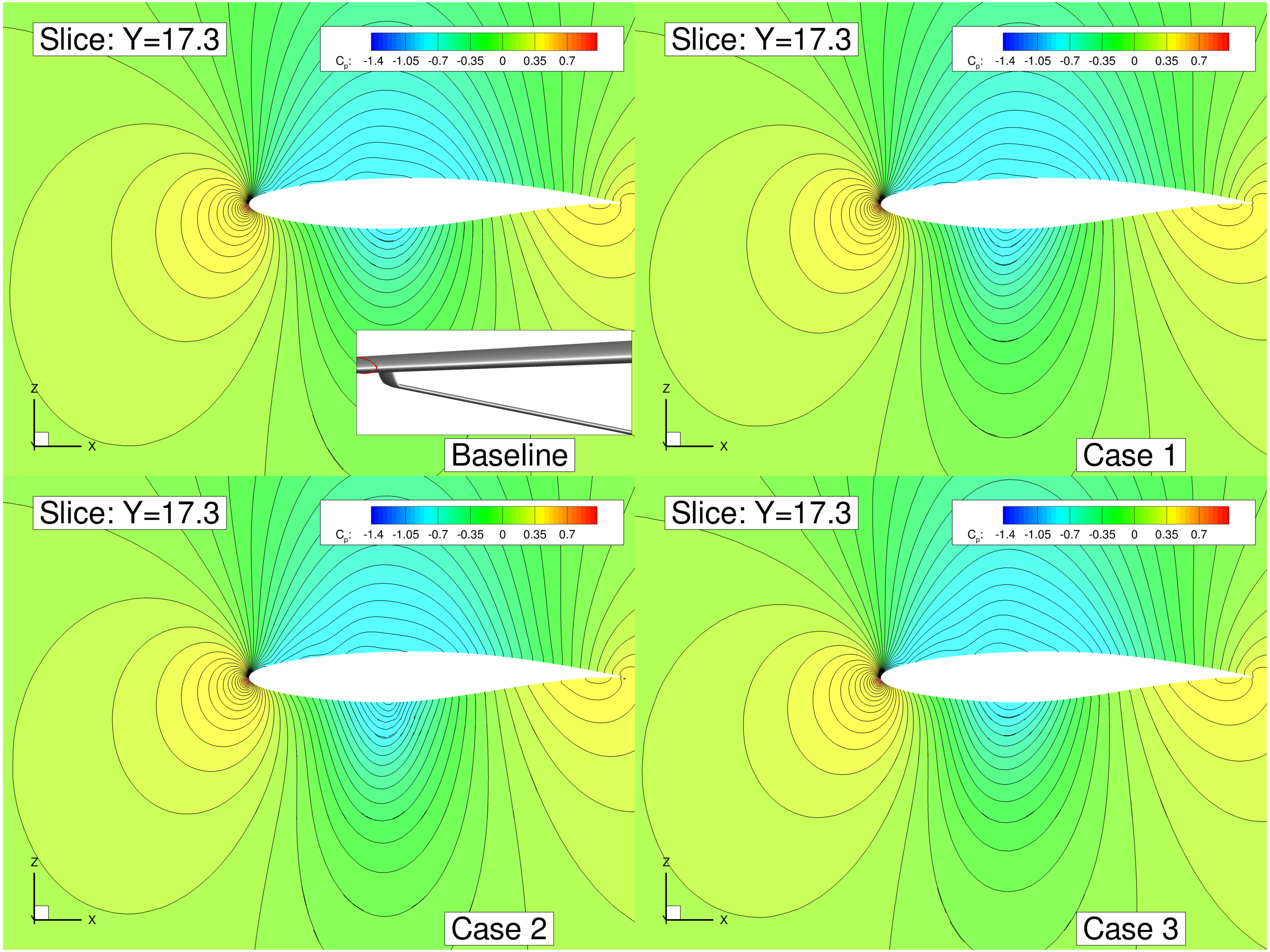

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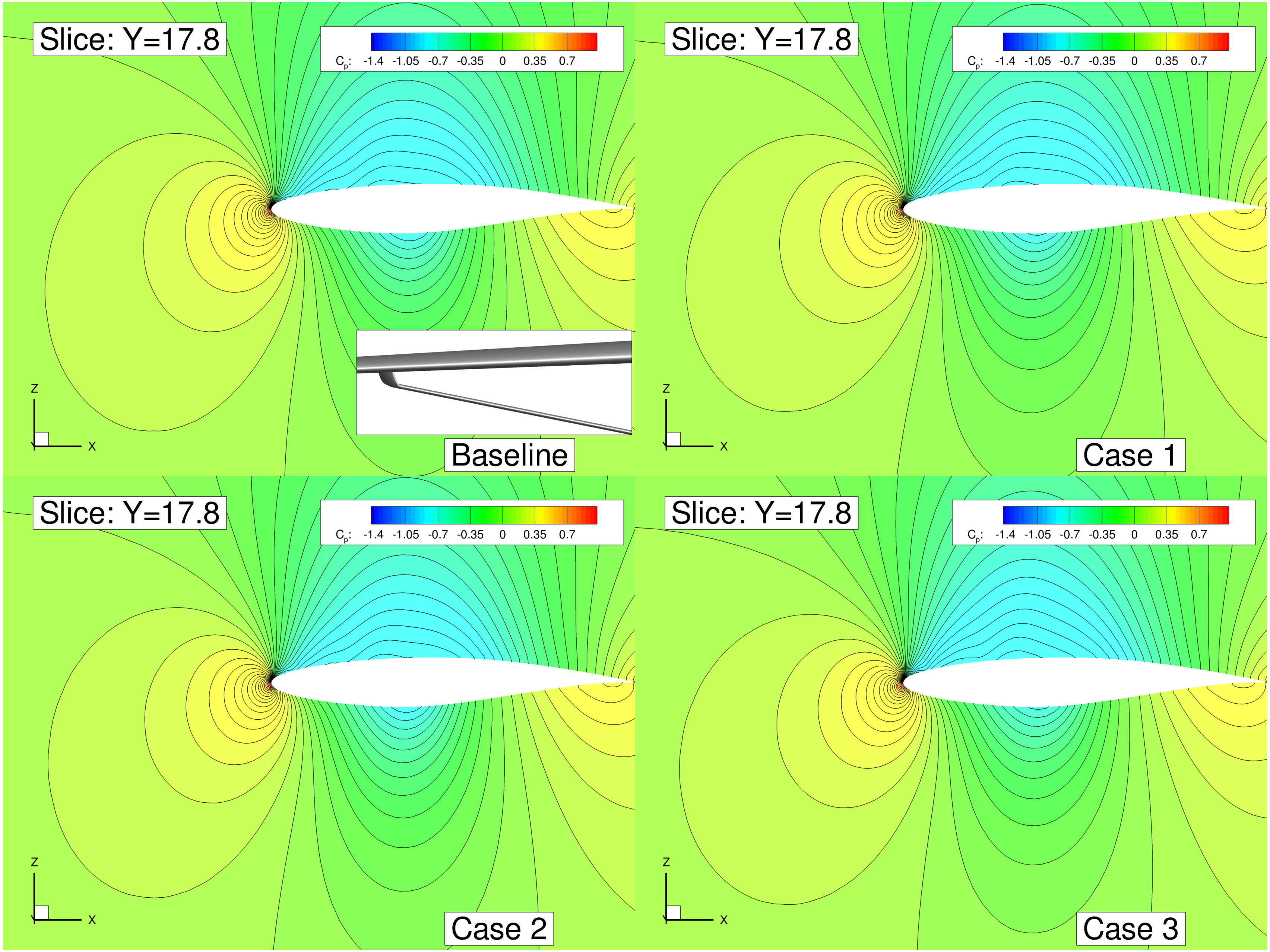

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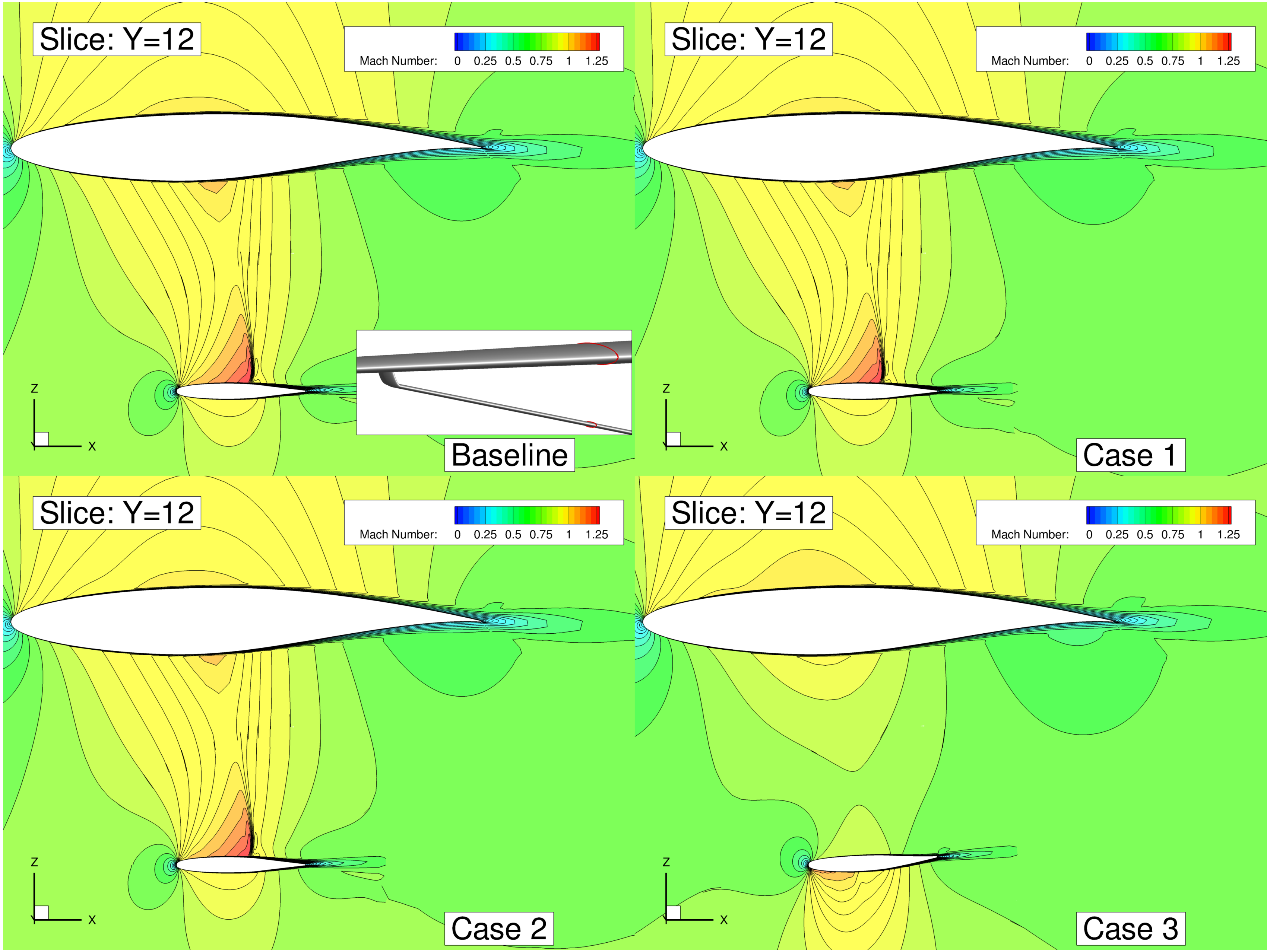

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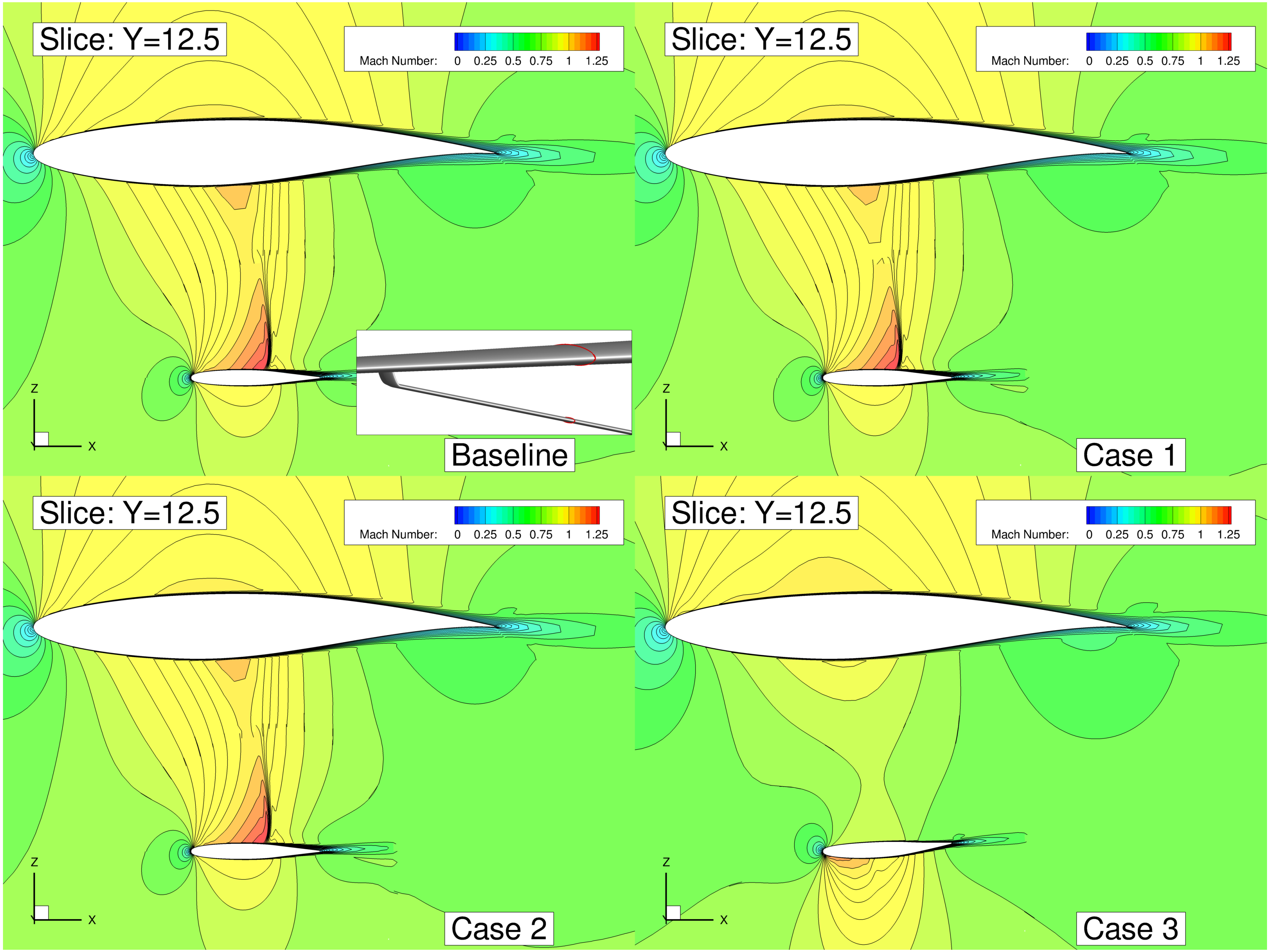

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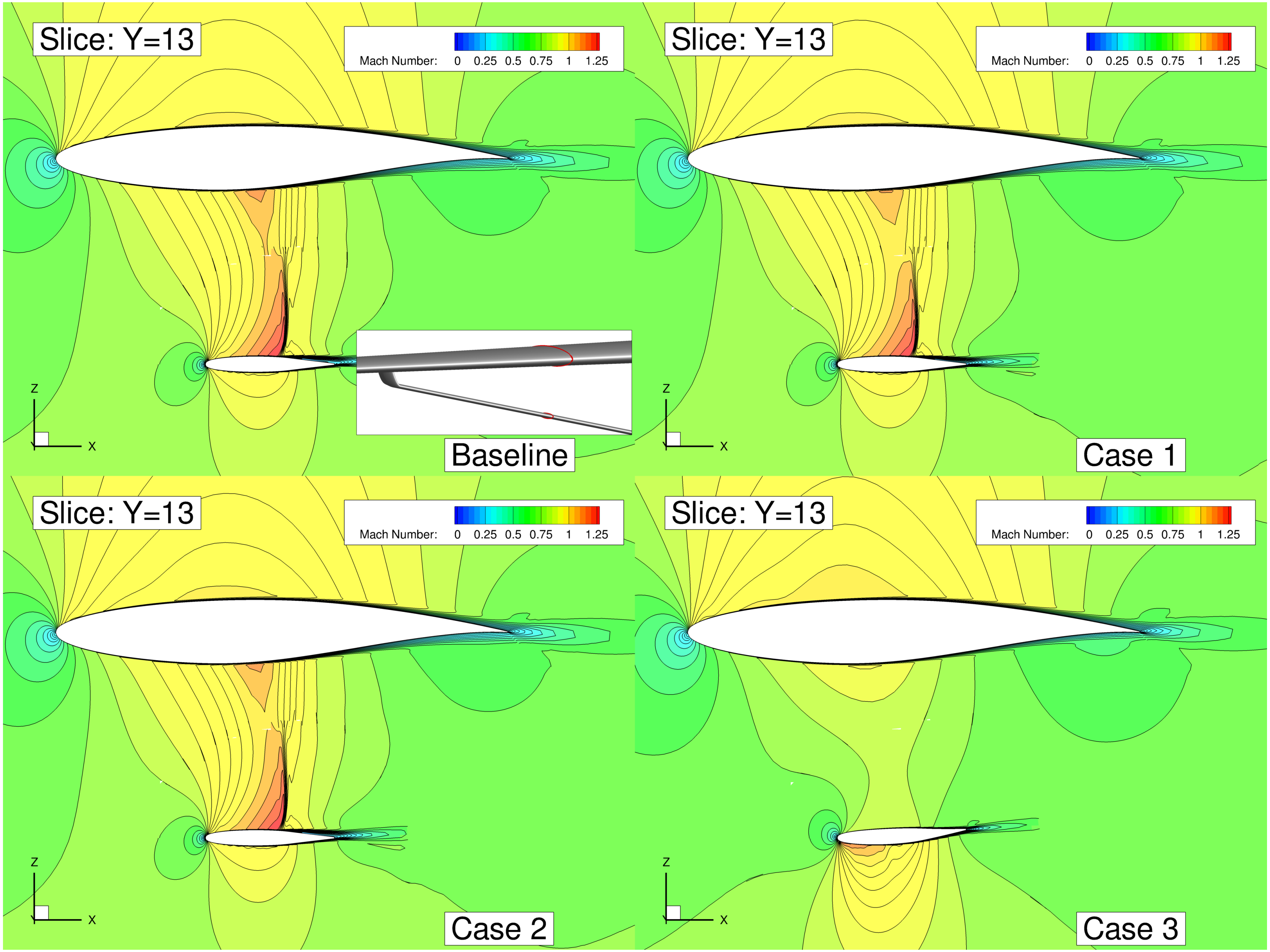

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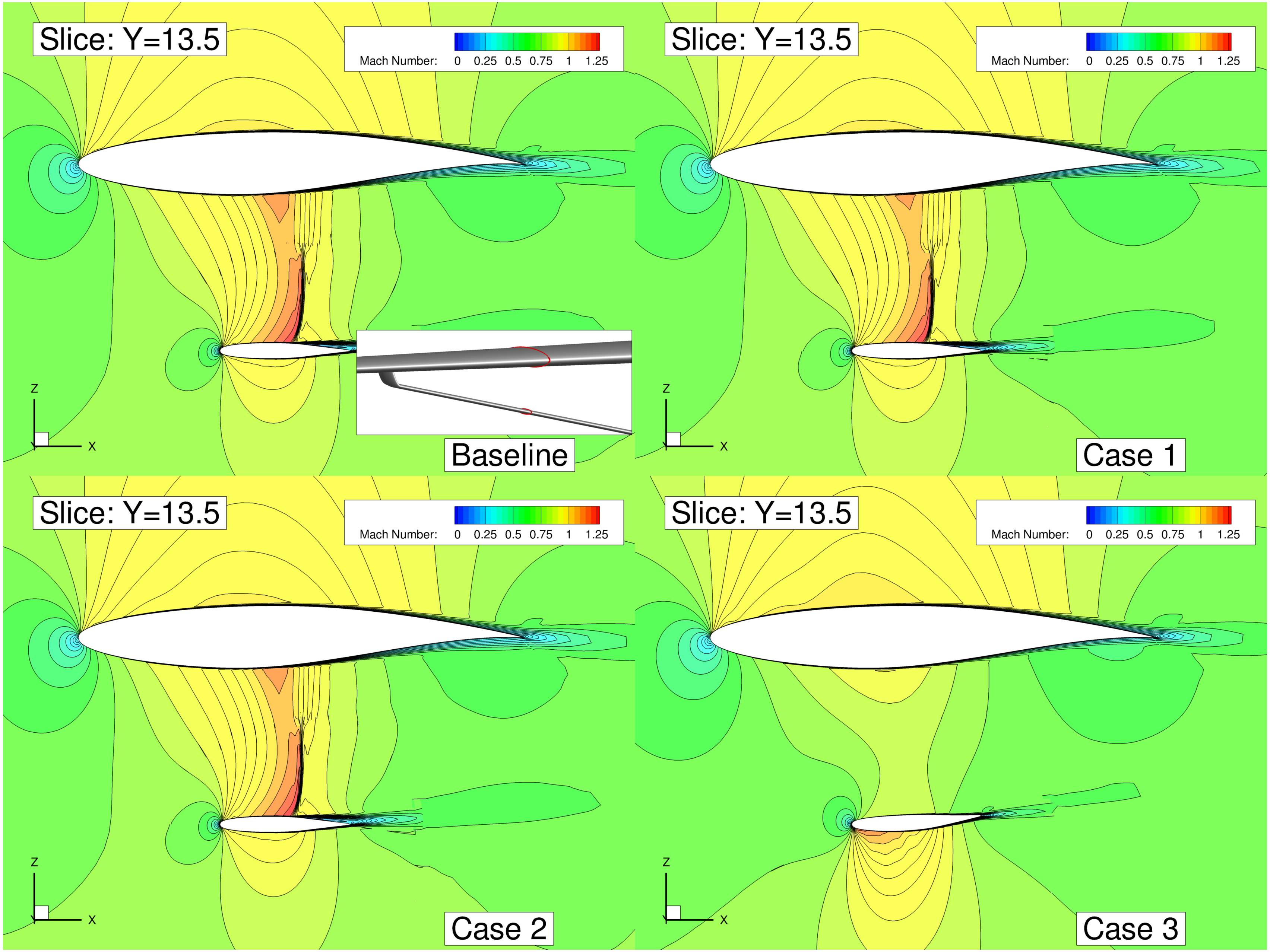

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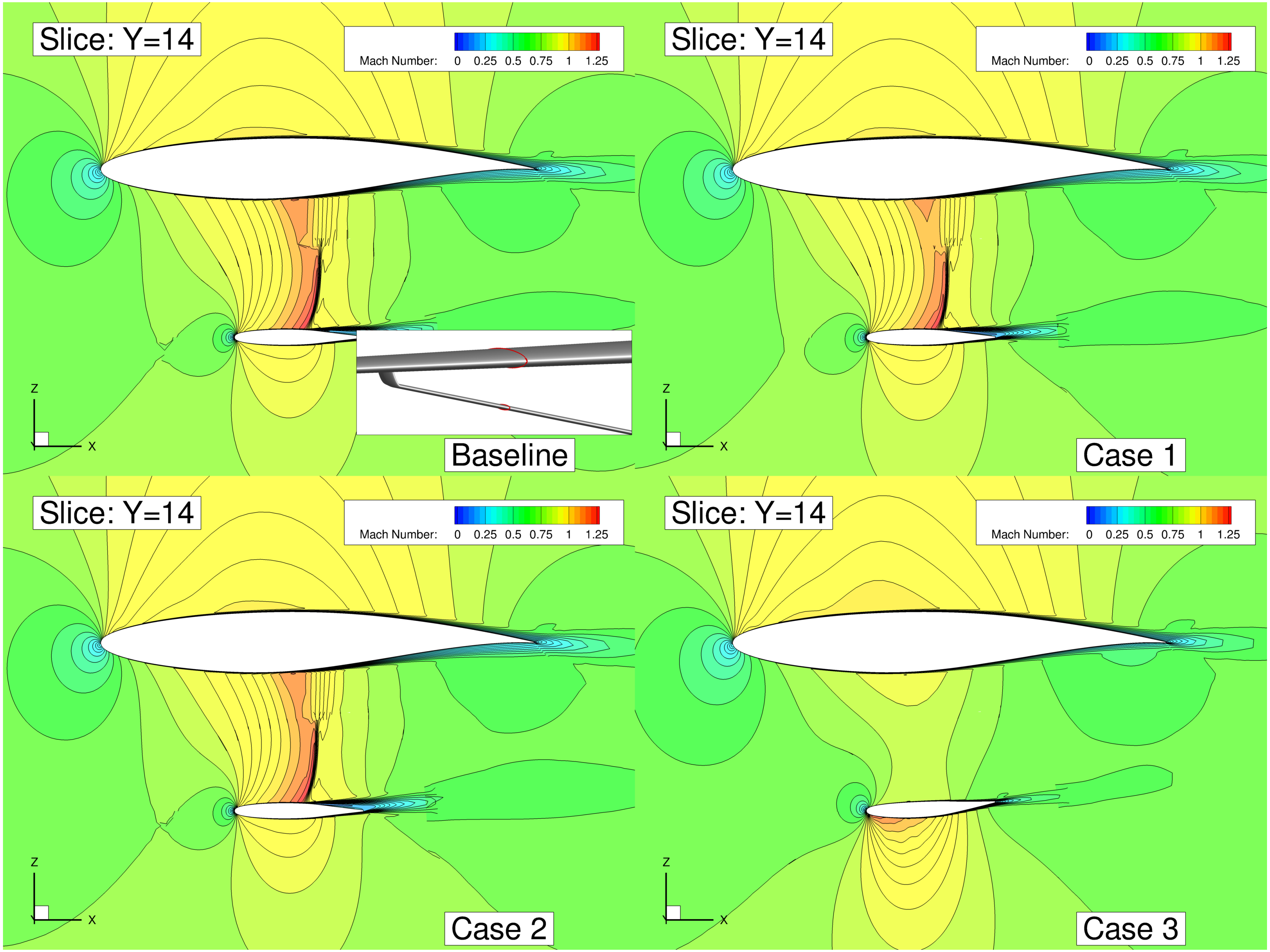

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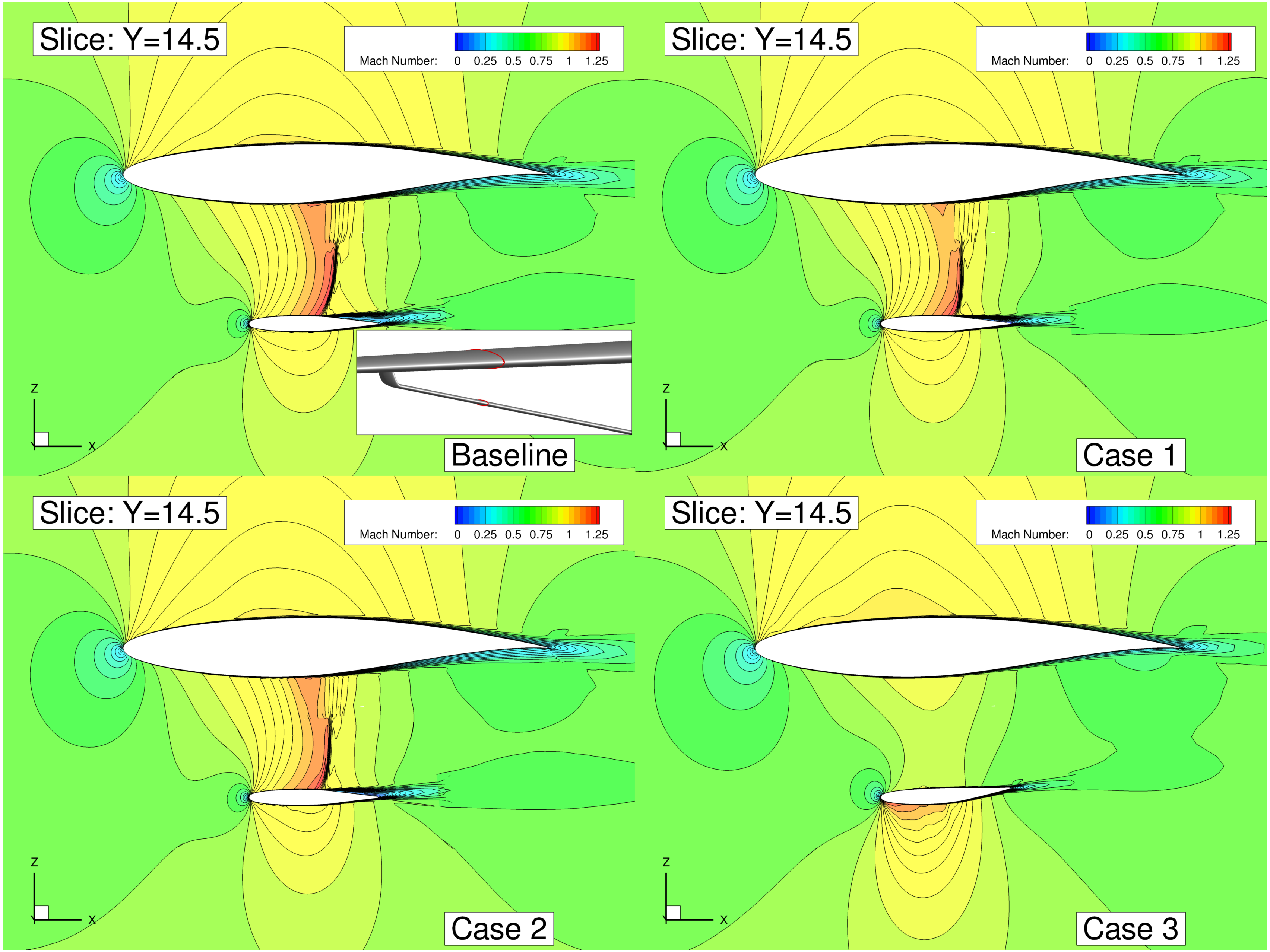

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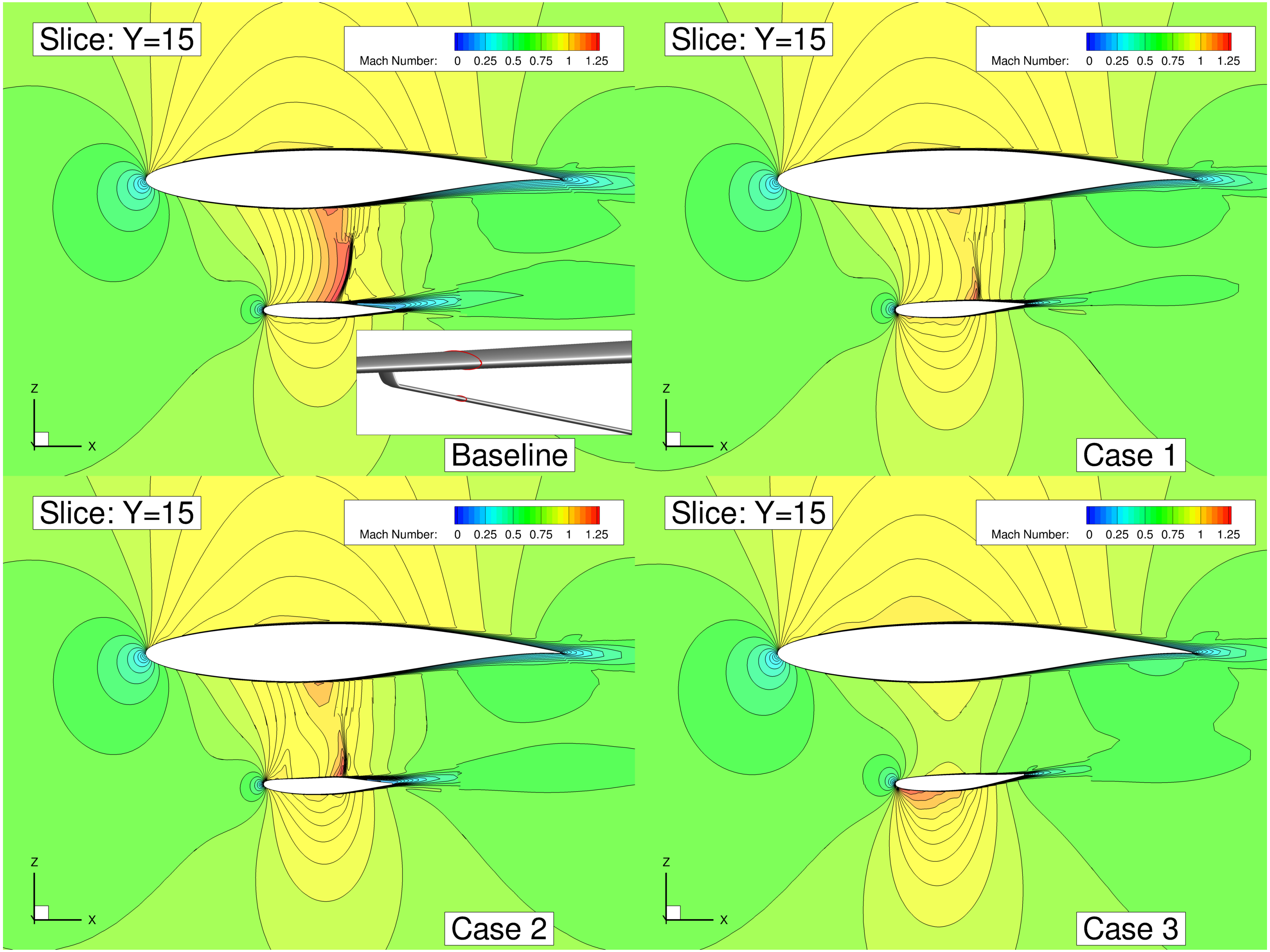

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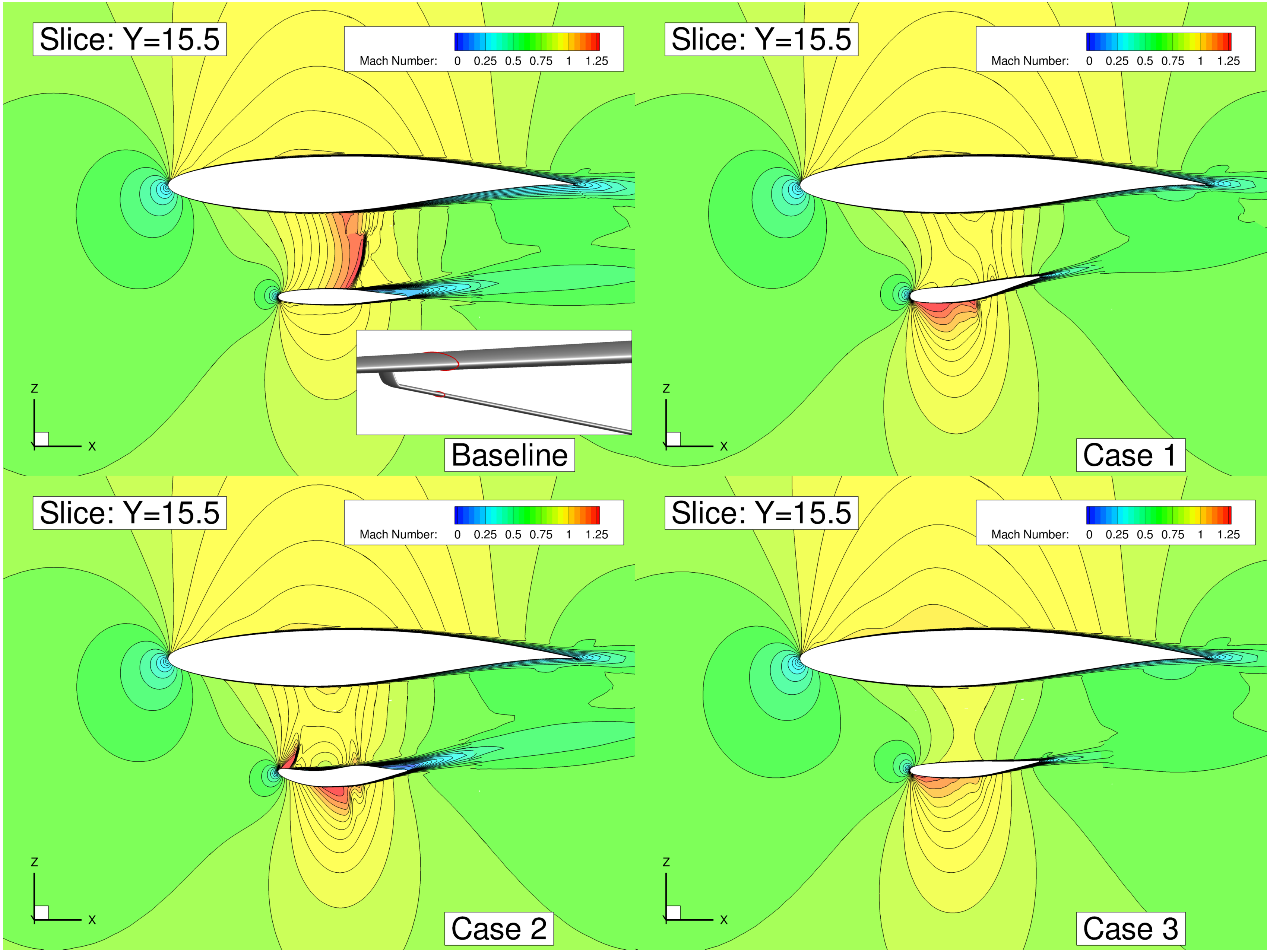

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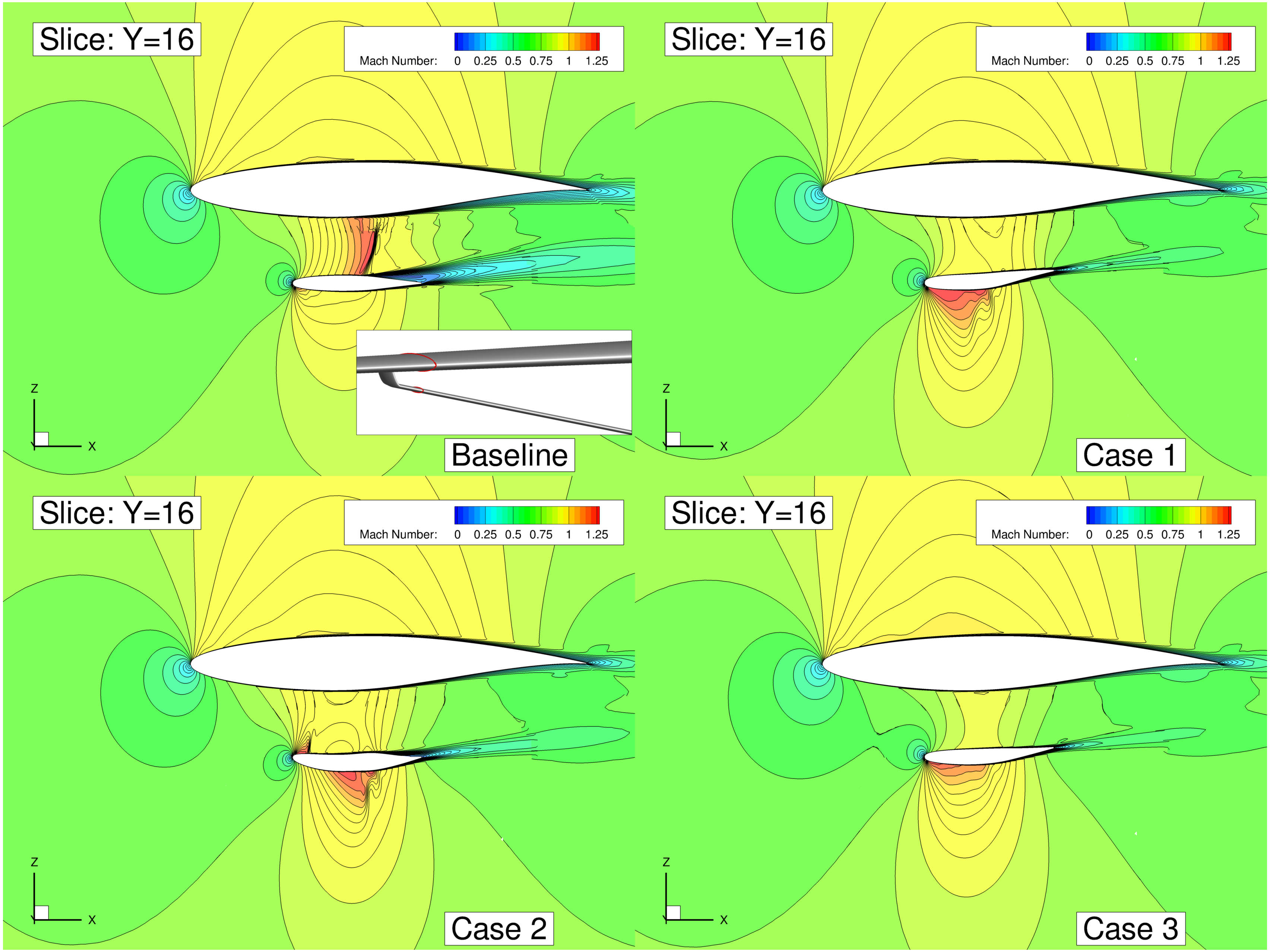

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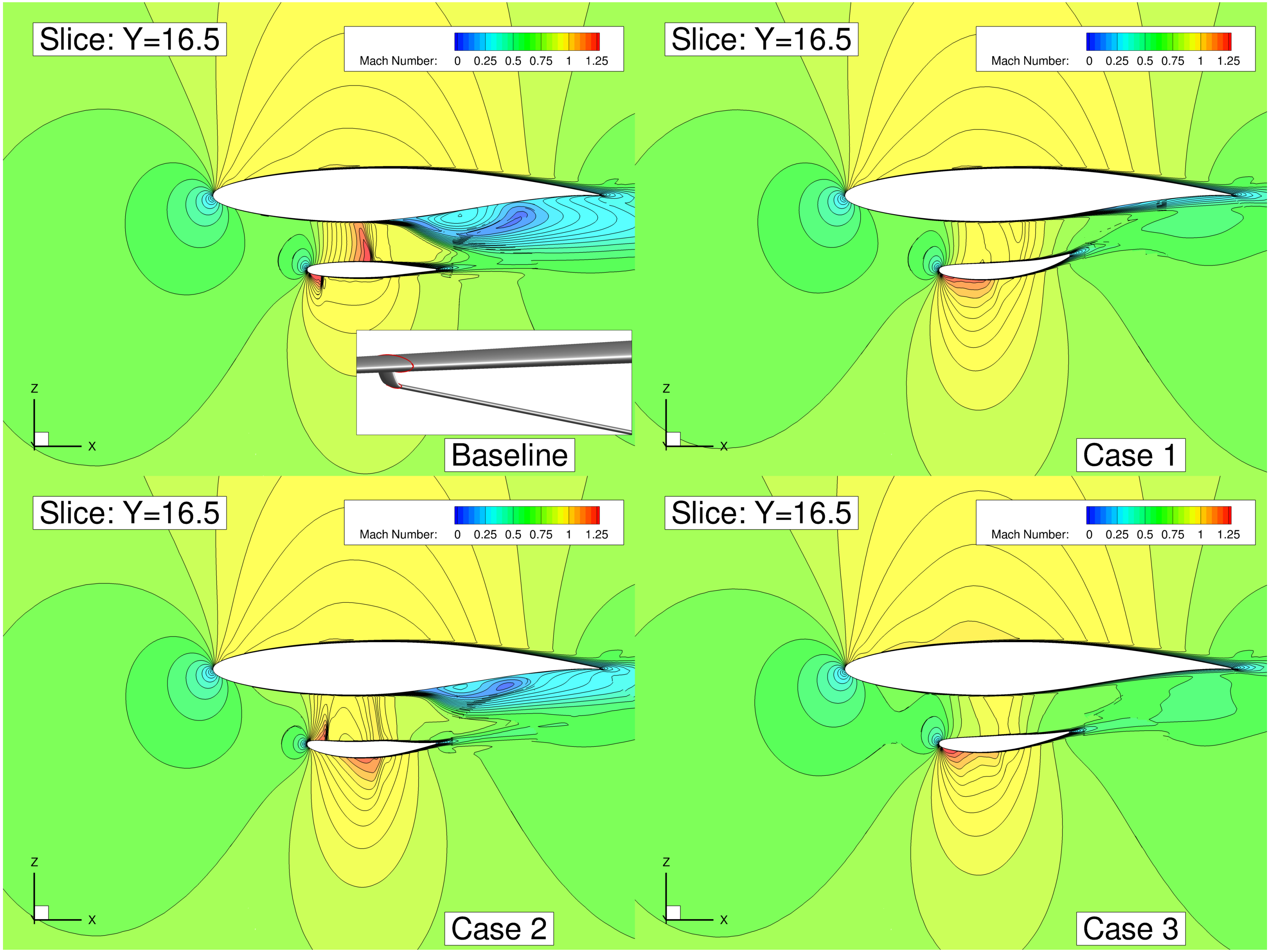

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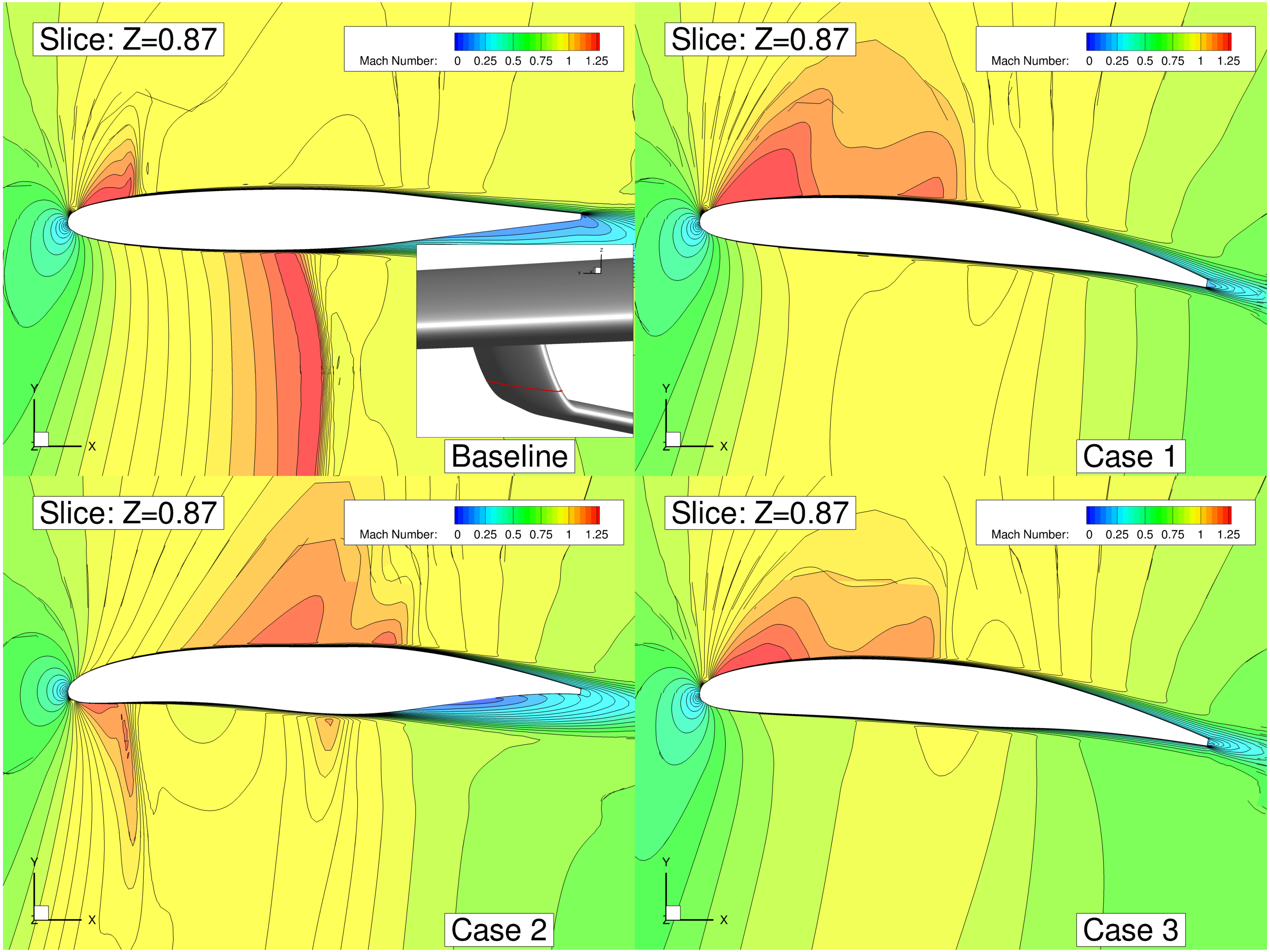

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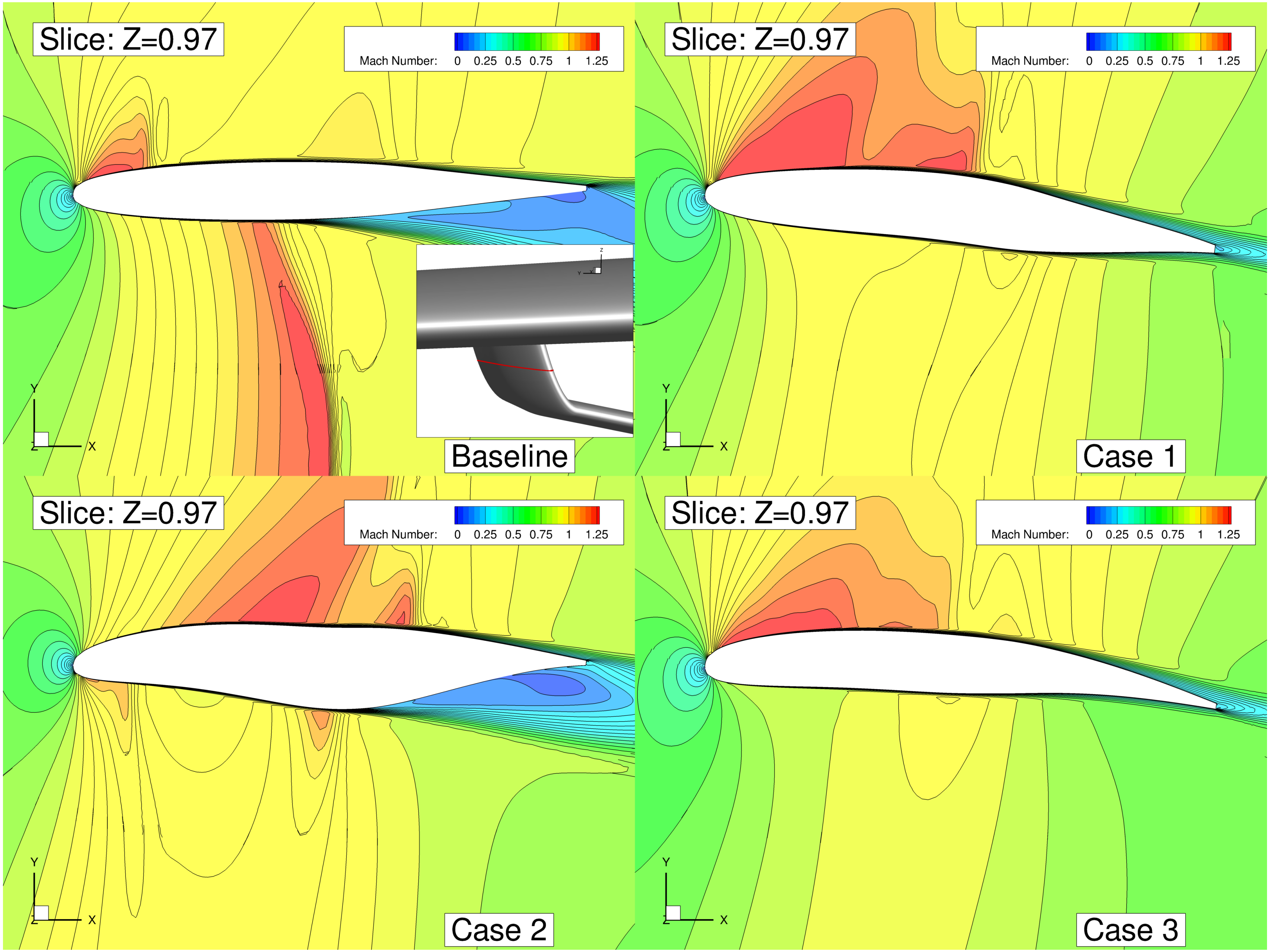

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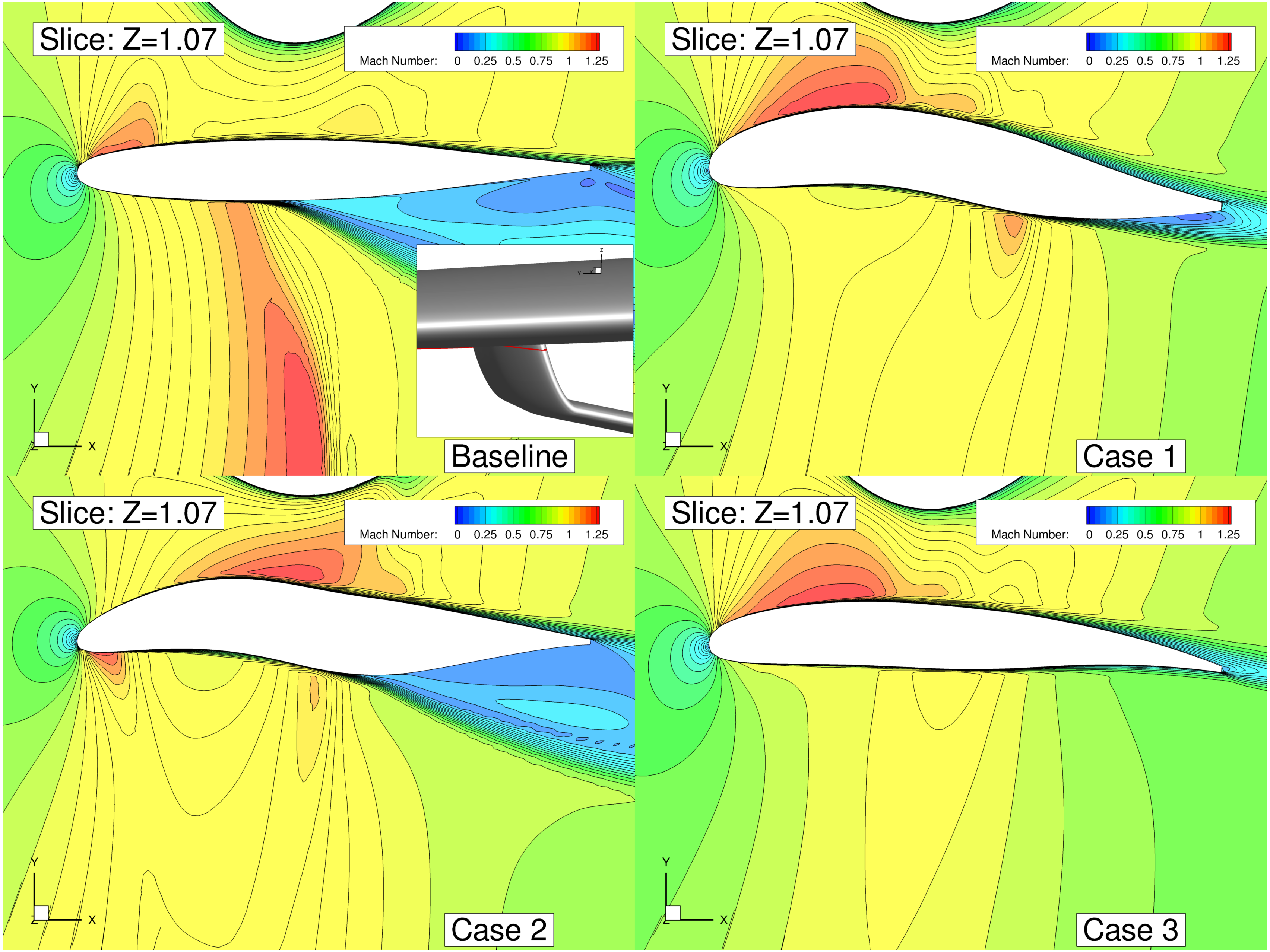

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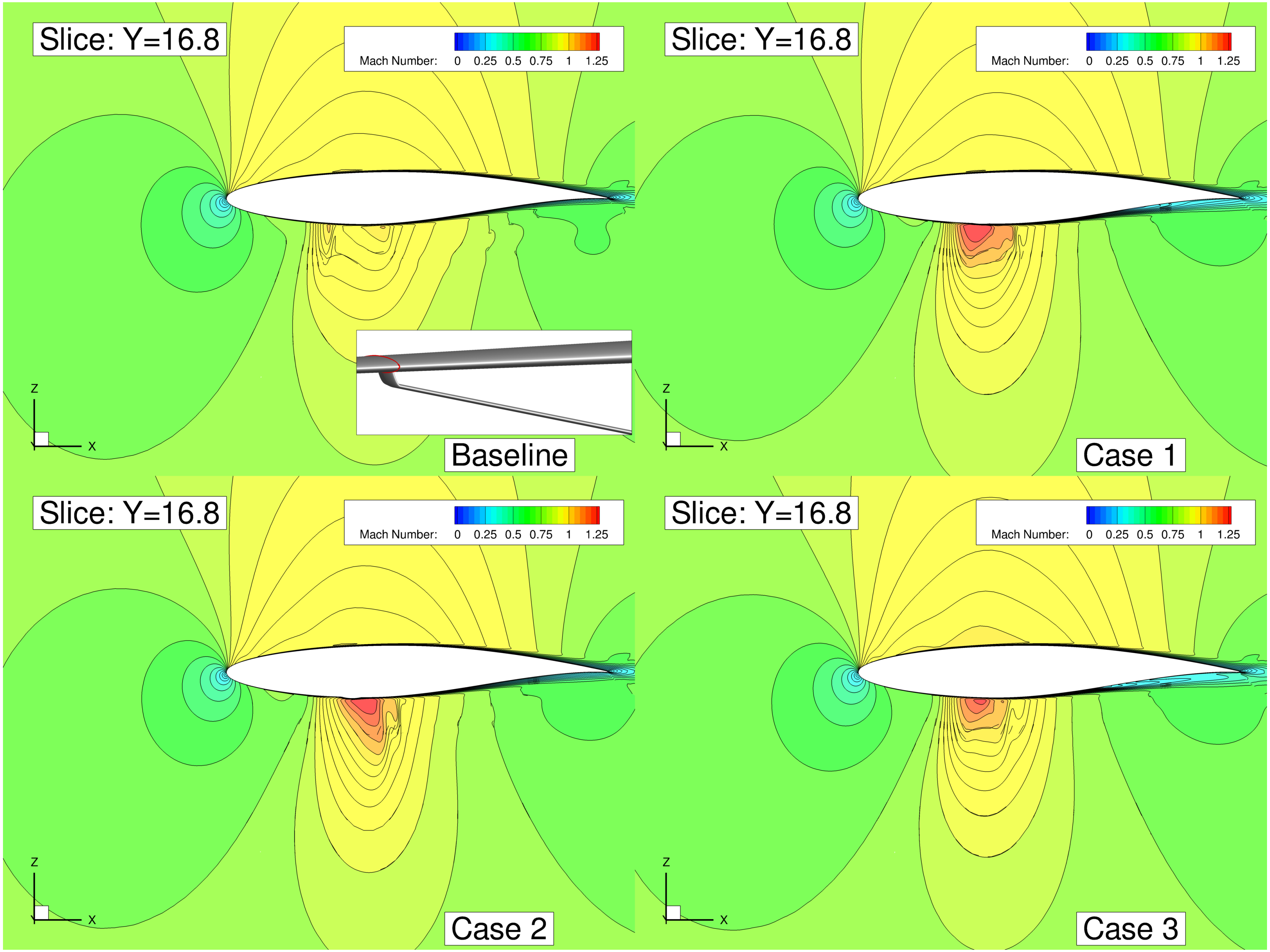

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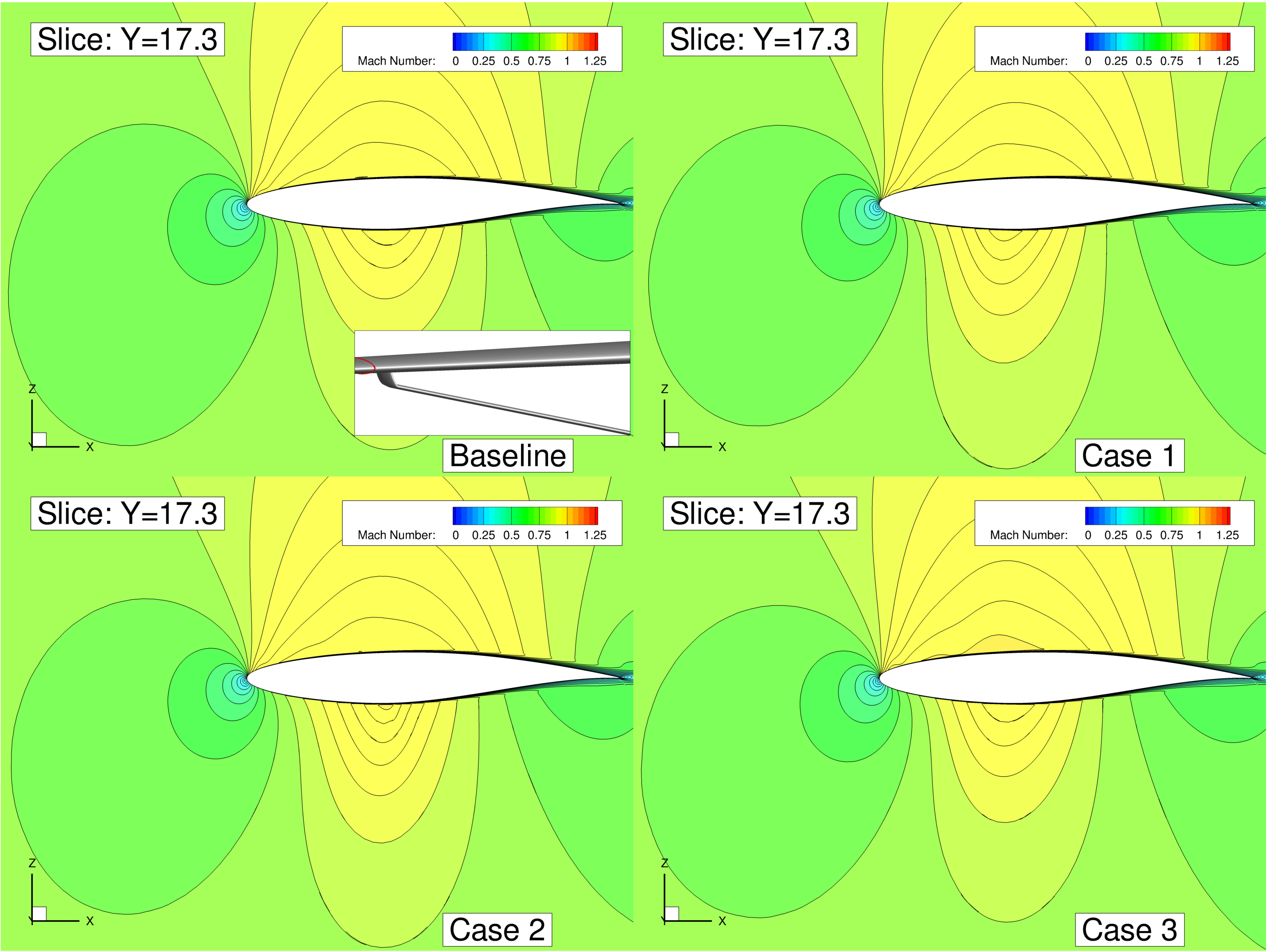

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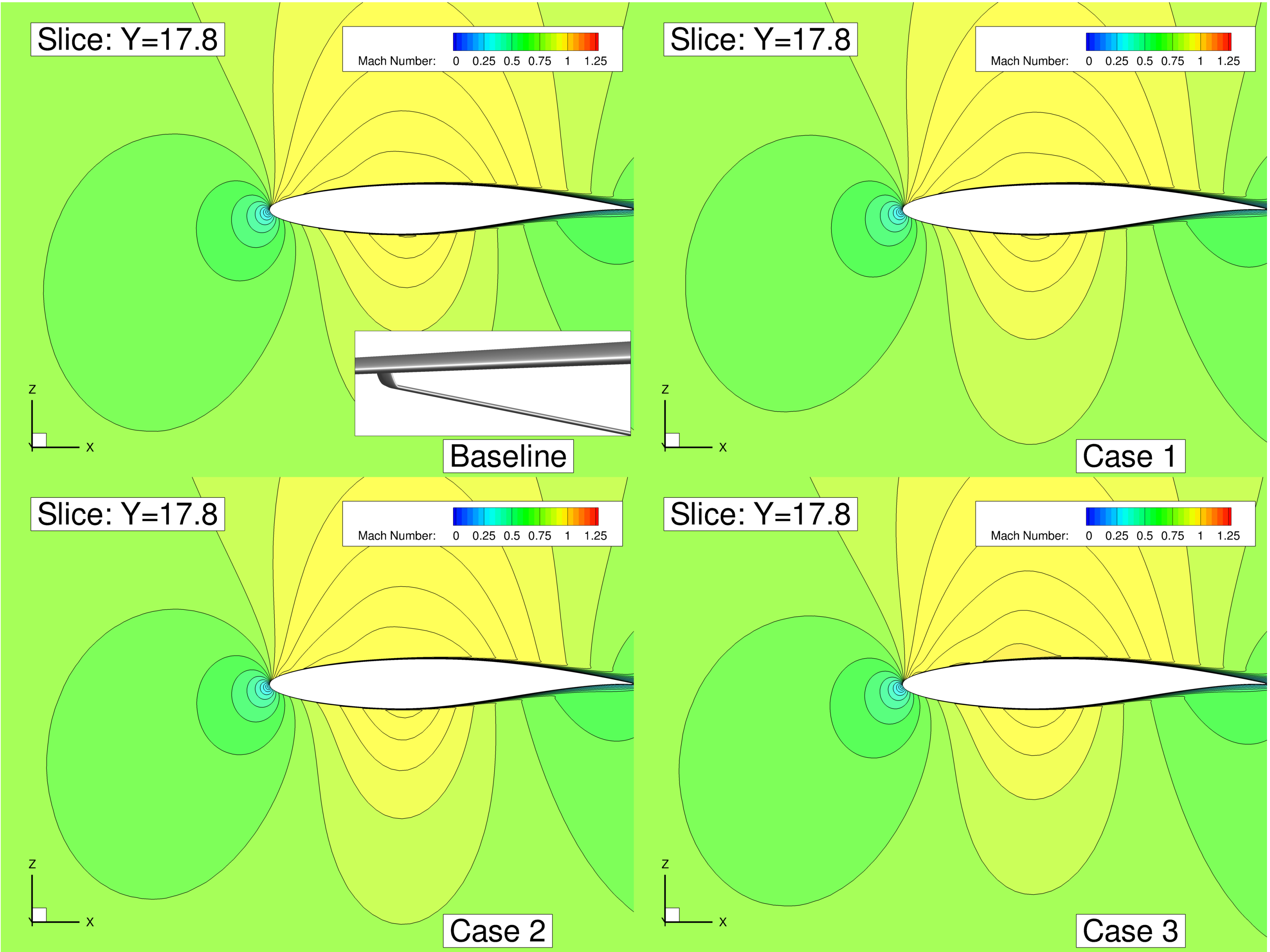
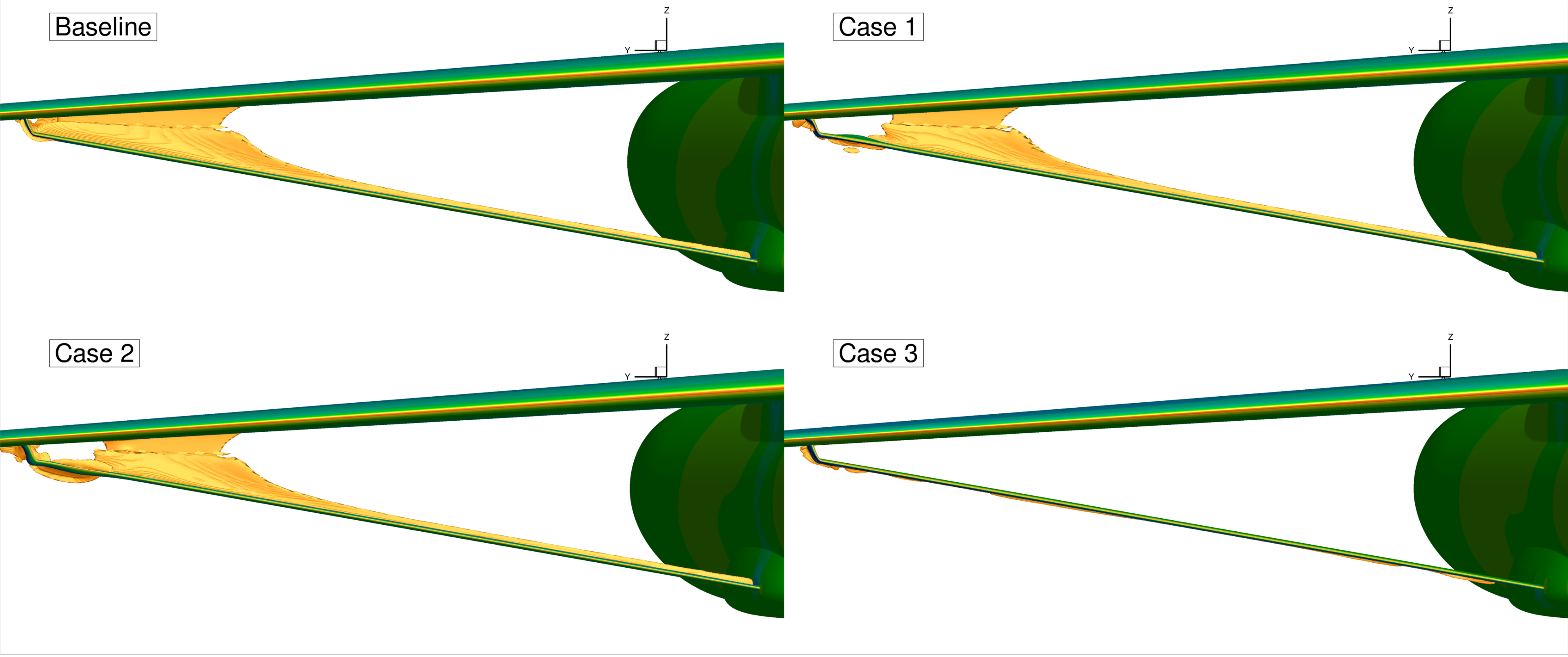


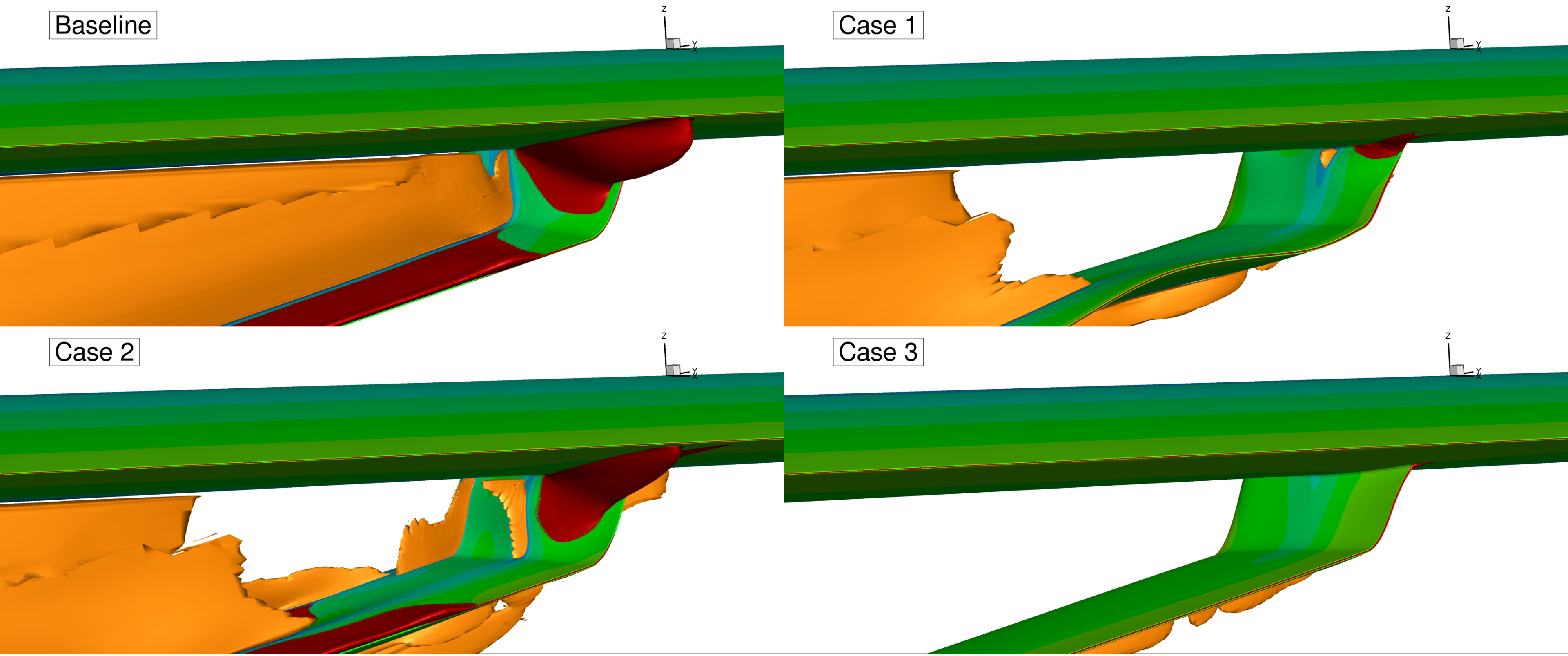
Shock Surface Visualization

- Case 1 successfully removes shock in design region
- Full truss redesign has weak shock on lower surface



Separated Flow

- All designs reduce the amount of separated flow at the strut-wing junction
- Red iso-contour at Vx=-.0001



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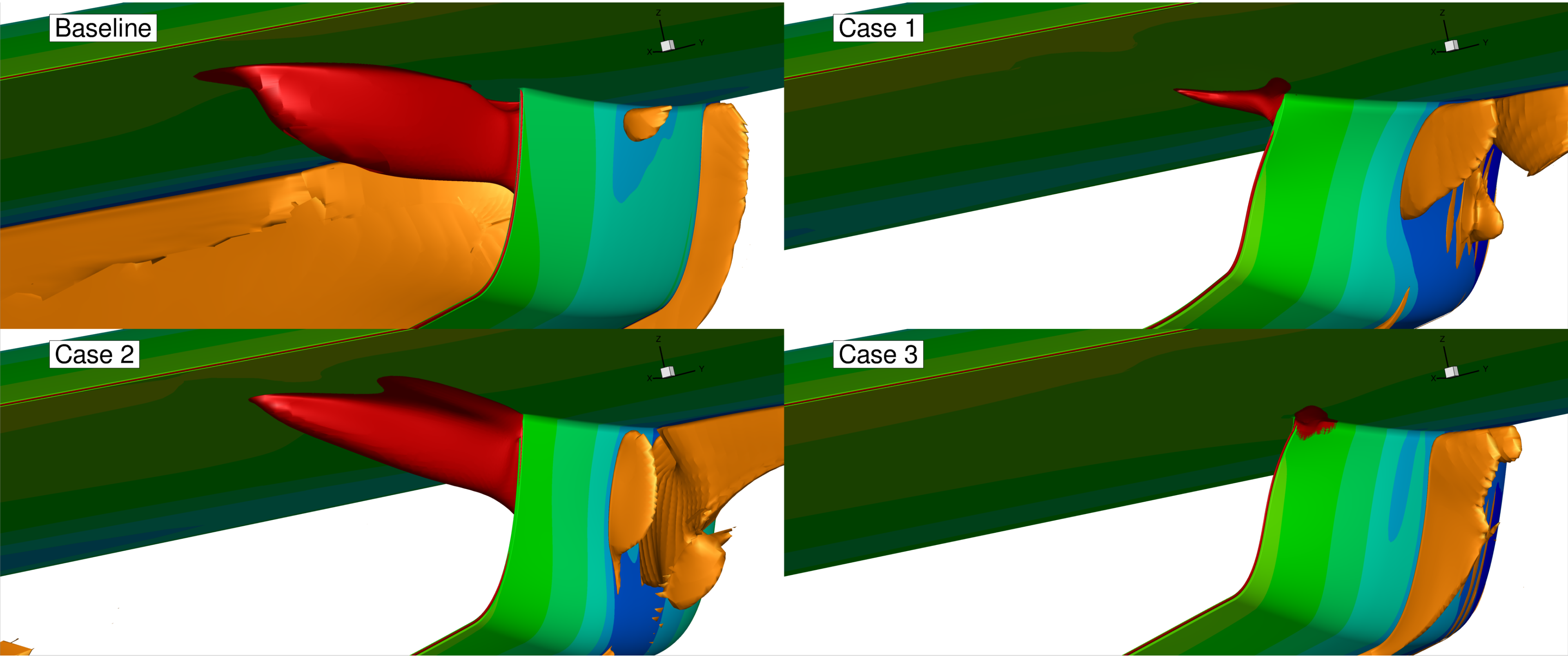
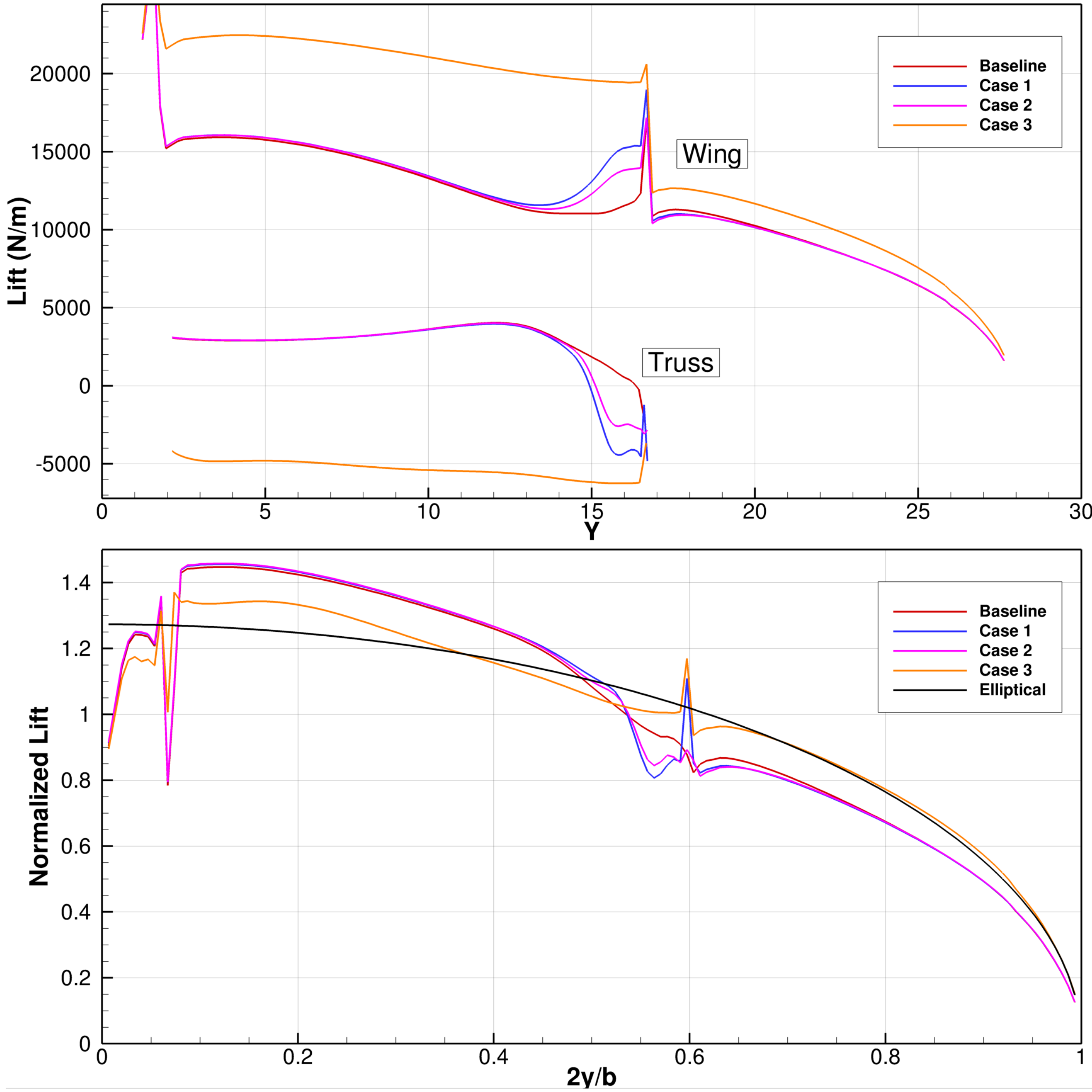


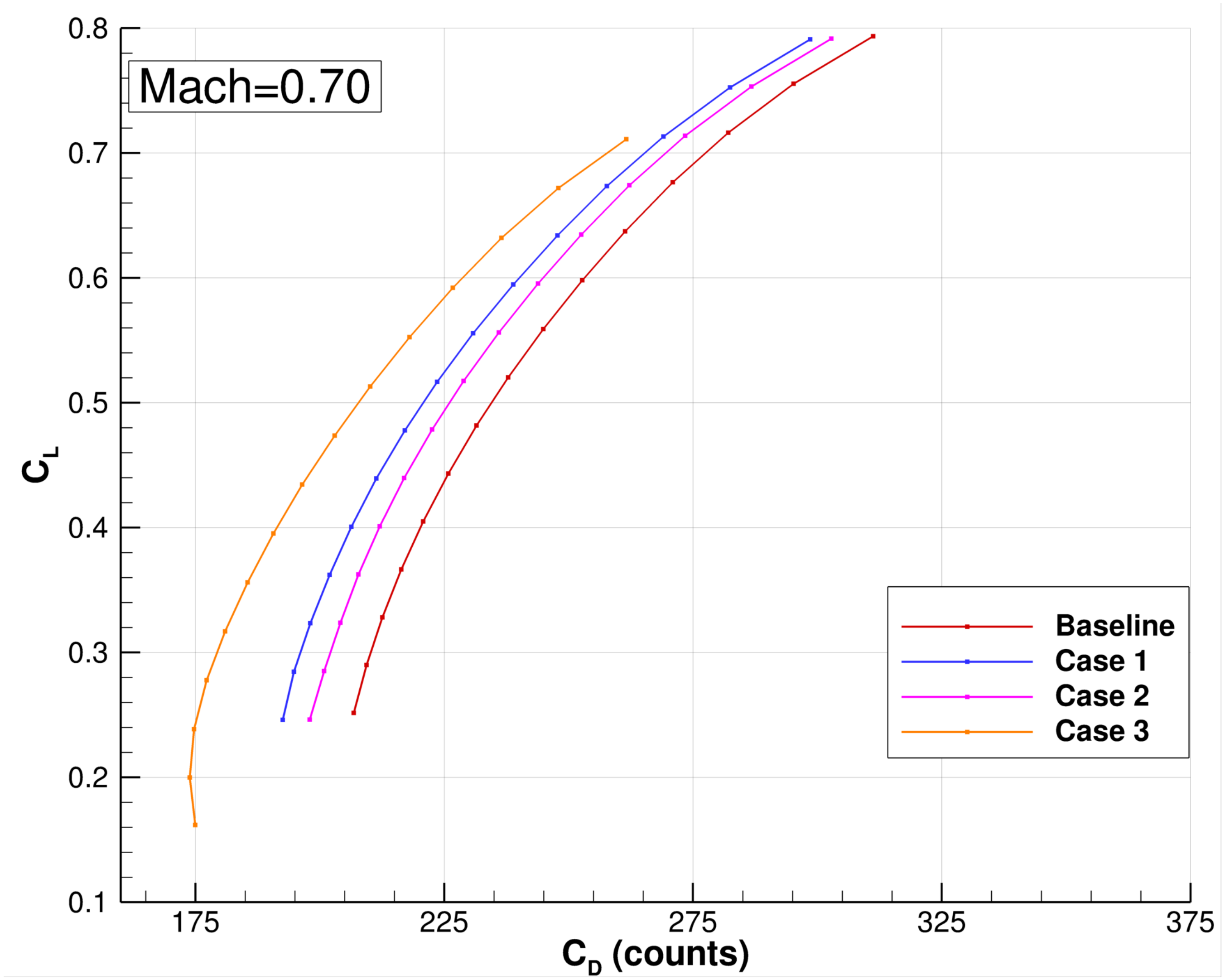
Lift Distributions

- All optimized designs reduce truss lift
- Nearly elliptical lift distribution and increased angle of attack for case 3
- Negative truss lift is optimal!



Off-Design Performance

- Consistent improvement across Mach and angle of attacks



Off-Design Performance

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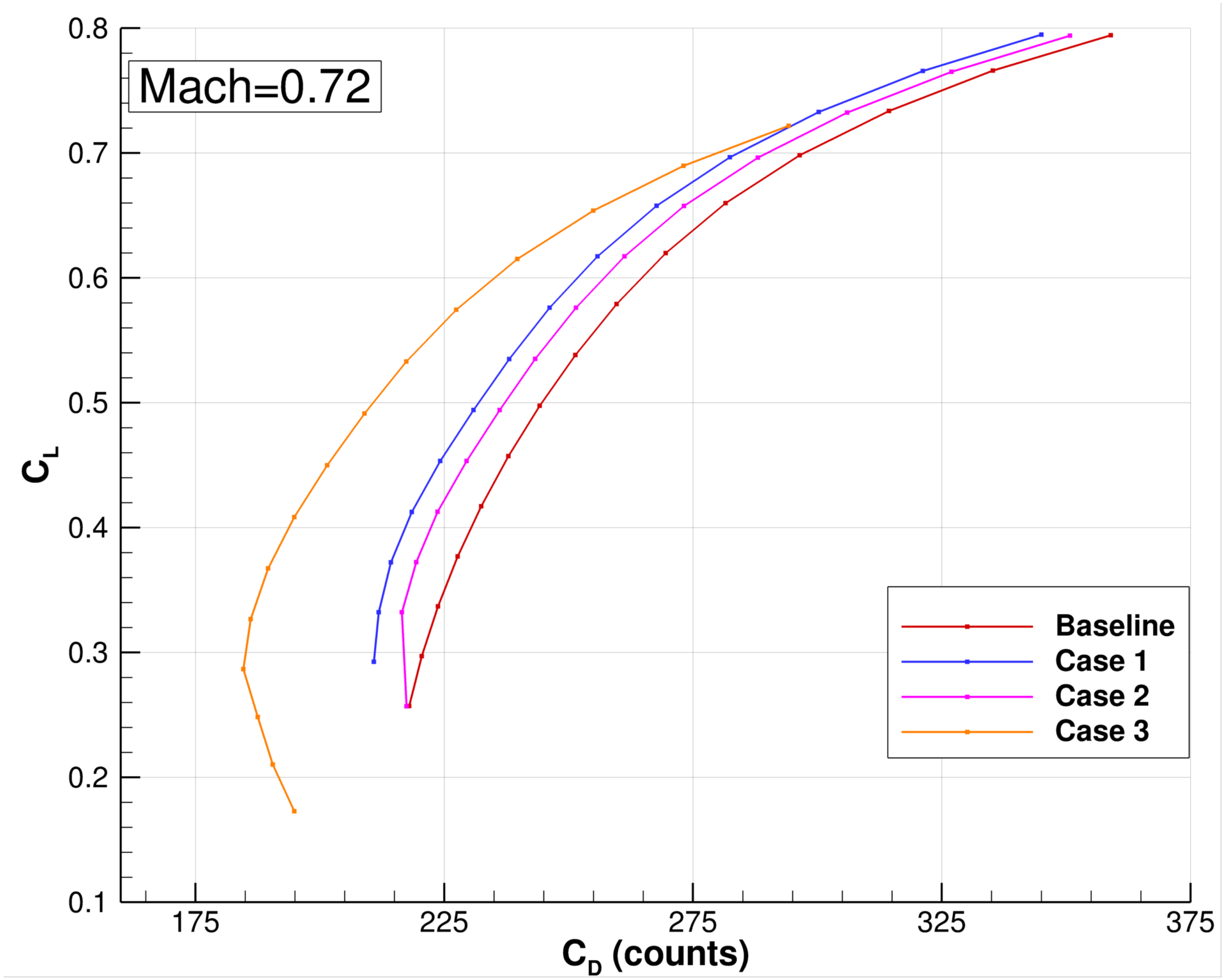

Off-Design Performance

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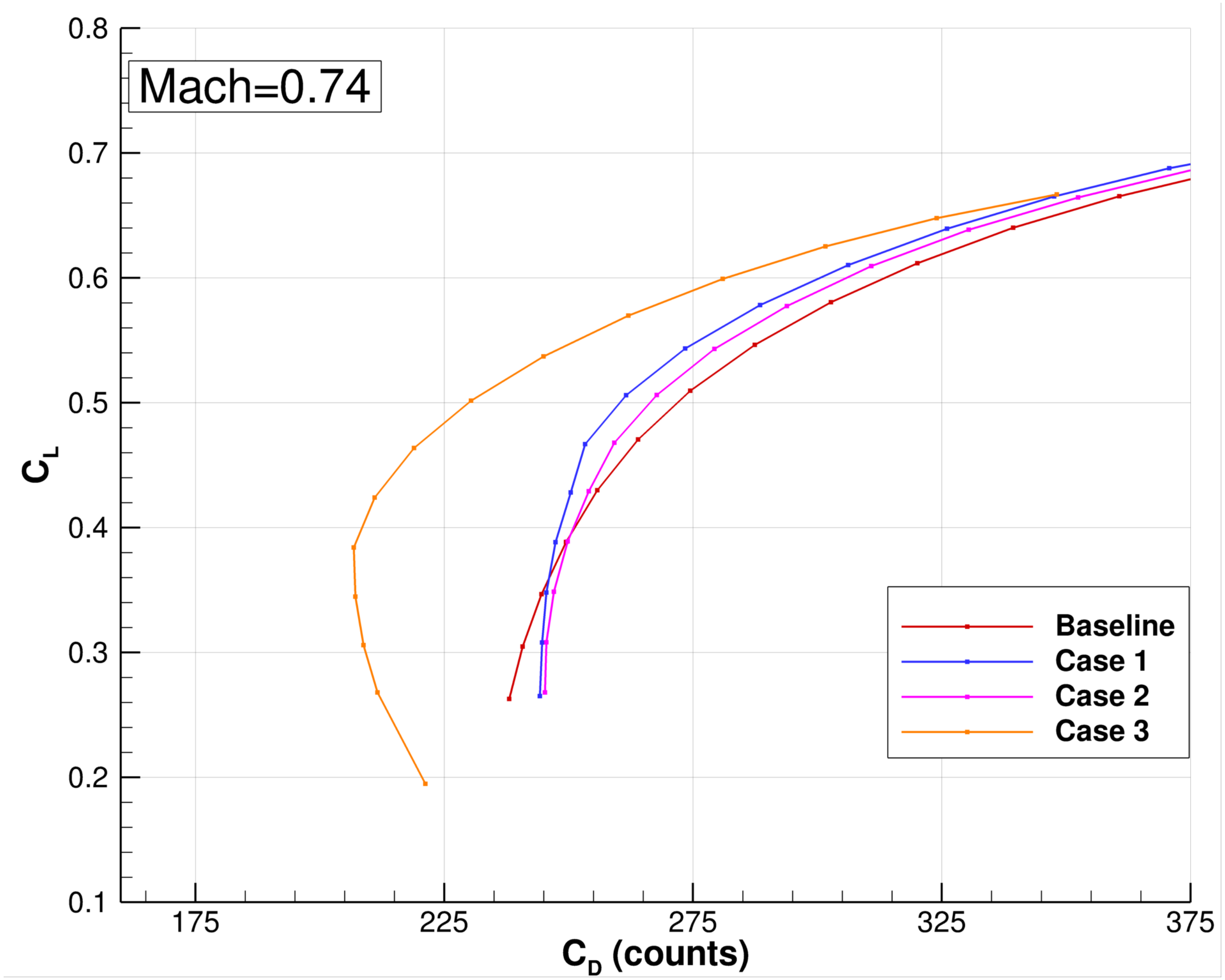
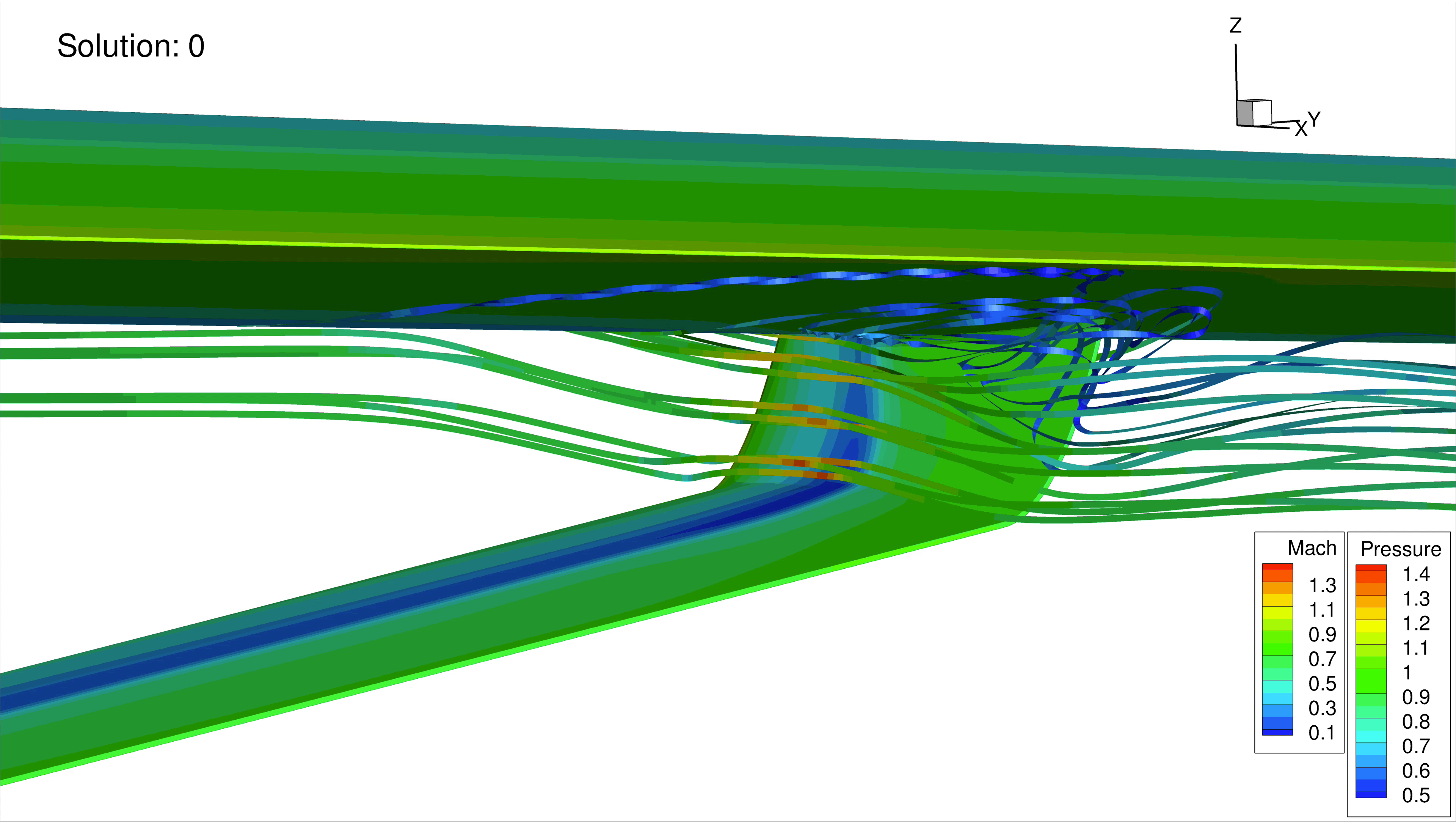
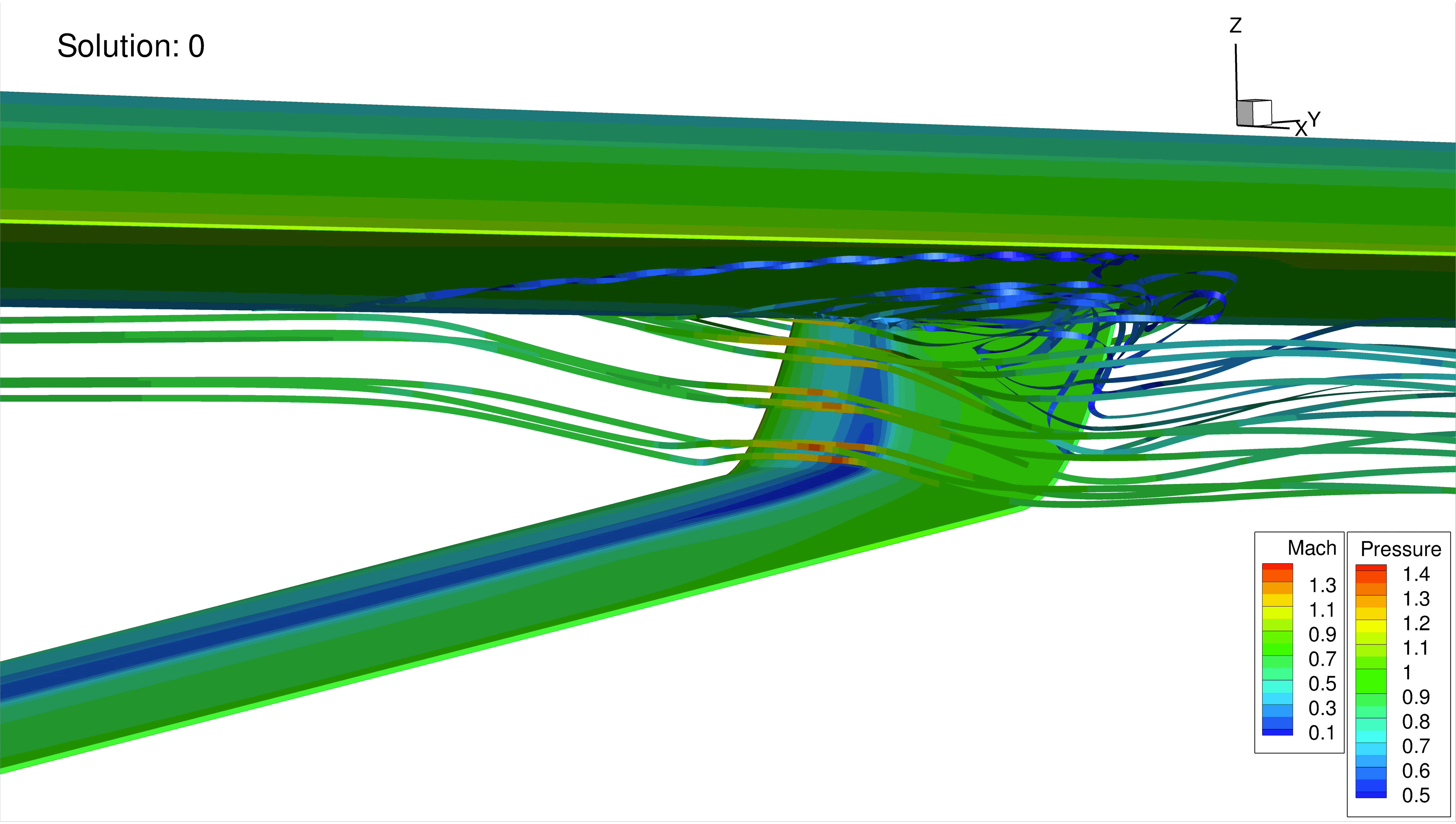


Optimization Case 1



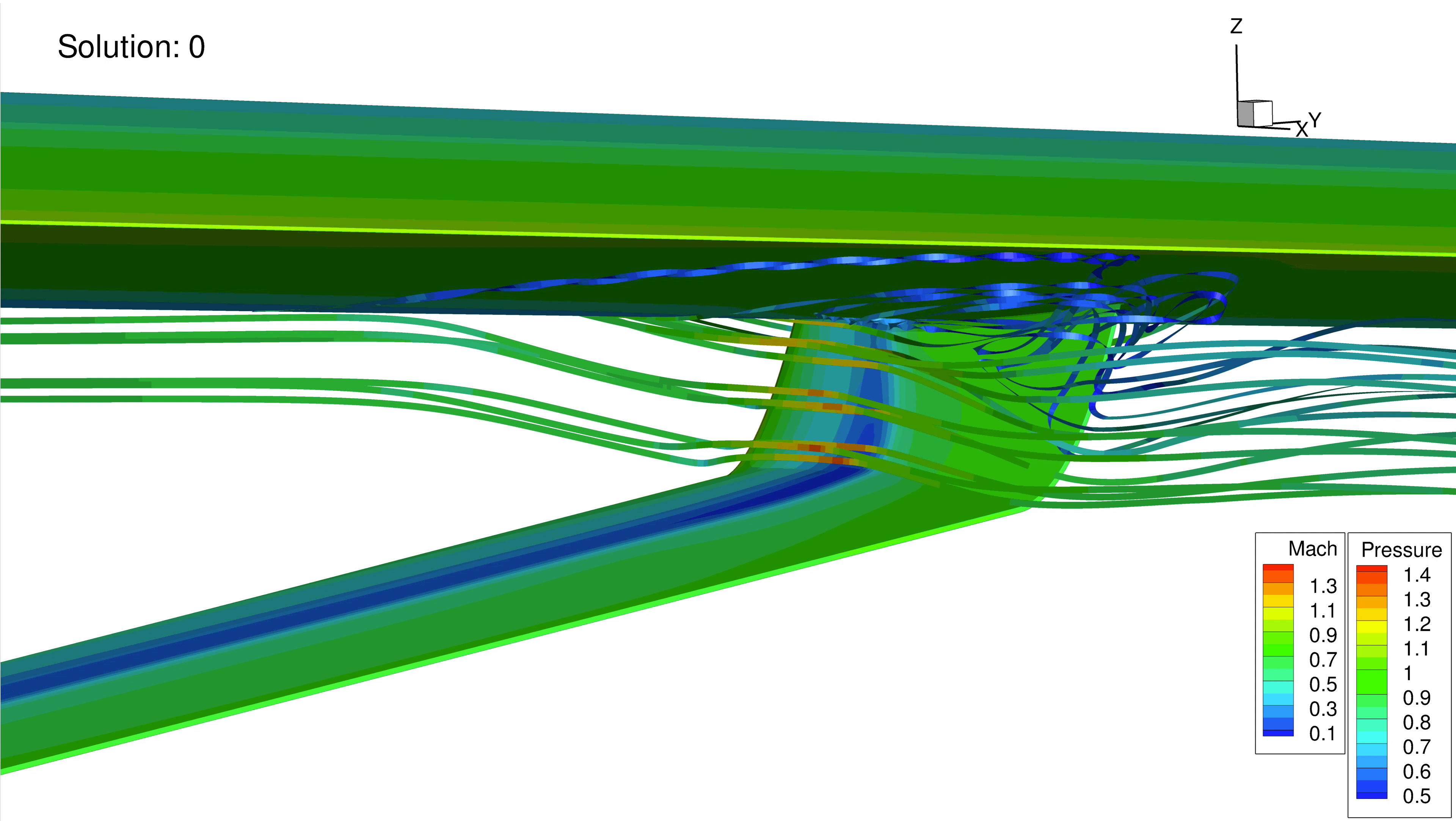
Pressure is shown on the surface. Stream ribbons are colored by Mach number.

Optimization Case 2



Pressure is shown on the surface. Stream ribbons are colored by Mach number.

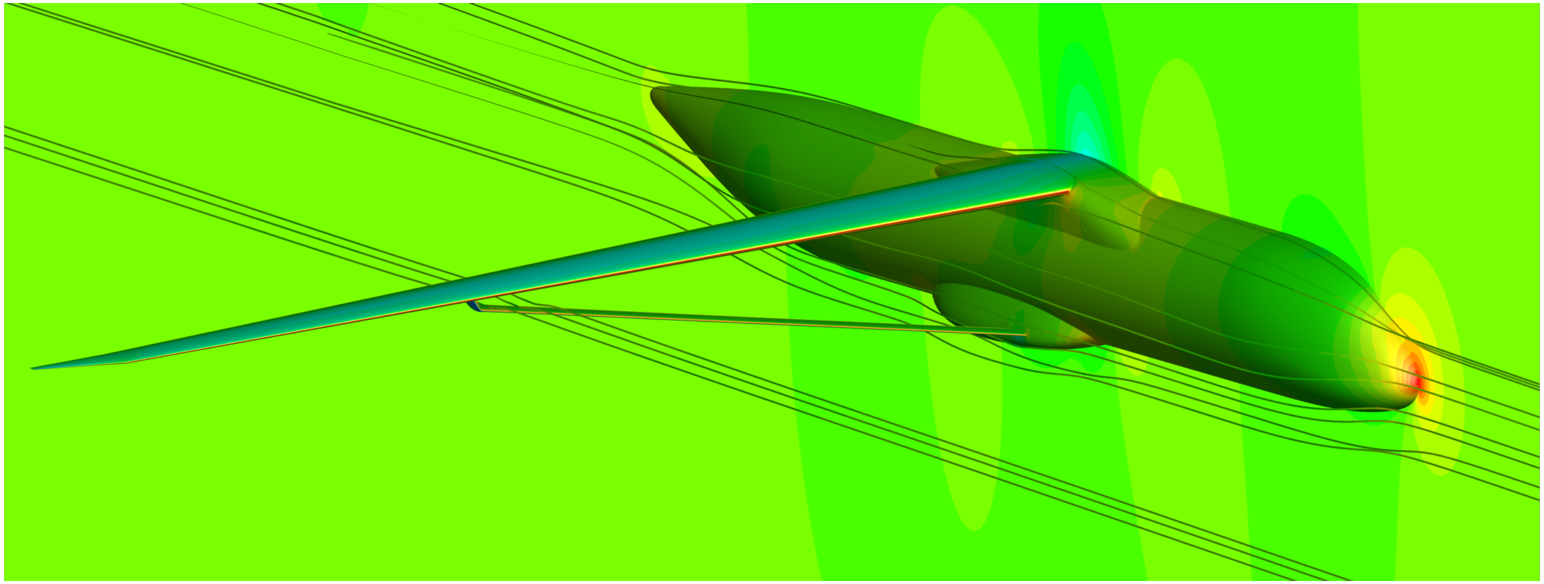
Optimization Case 3



Pressure is shown on the surface. Stream ribbons are colored by Mach number.

Summary

- Successfully redesigned truss-junction intersection
- Fast optimization turn-around times of under 2 hours
- 13.5 drag count reduction for Case 1
- 33.5 drag count reduction for Case 3
- In transonic flow, truss may have negative lift
- No cost associated with flow control device other than initial development costs
- Future work should include aero-structural trade-offs



Questions

This work is funded by Nasa Advanced Air Transport Technology (AATT), sub project High Aspect Ratio Wing (HAW)