

Fiber Optics Instrumentation Development



Patrick Hon Man Chan, PhD.
Allen R. Parker, Jr
W. Lance Richards, PhD.

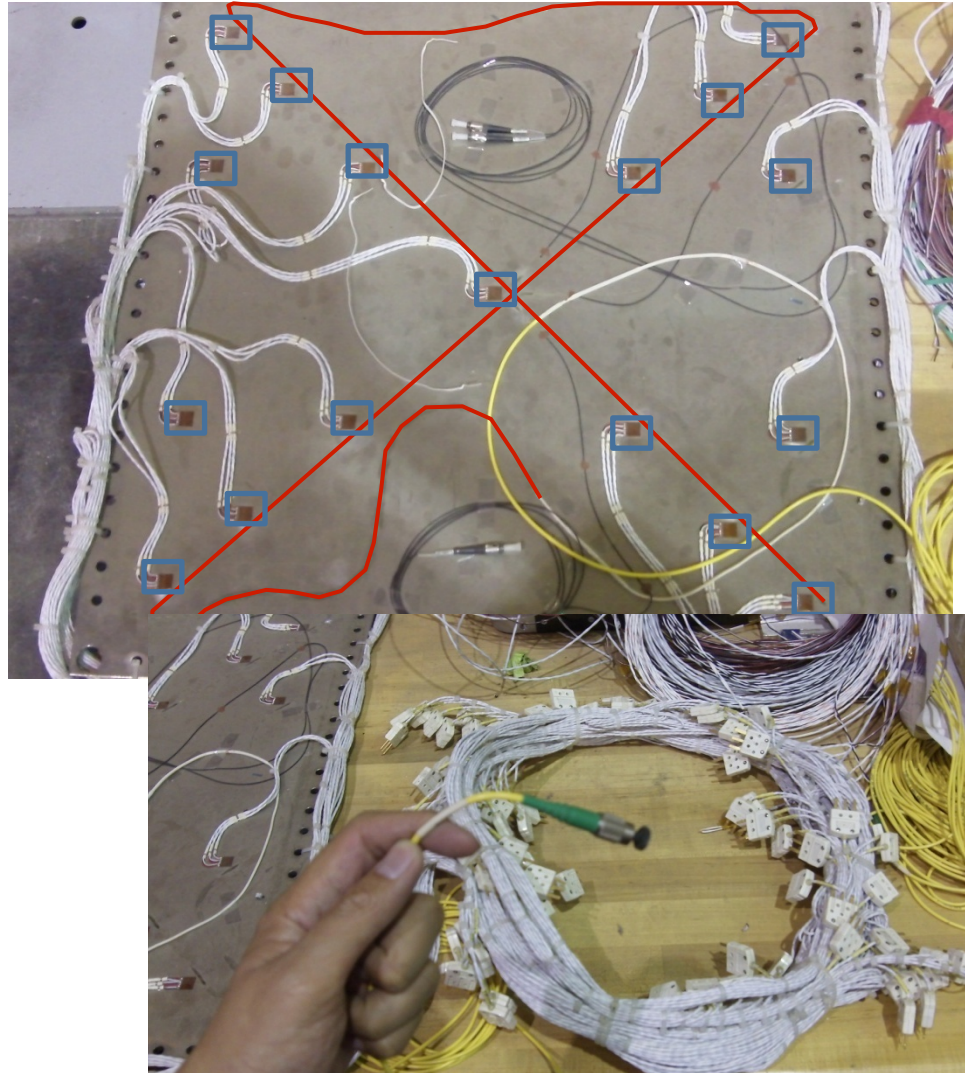
International Test and Evaluation
Association (ITEA) Meeting
July 21st, 2010



National Aeronautics and Space Administration
Dryden Flight Research Center

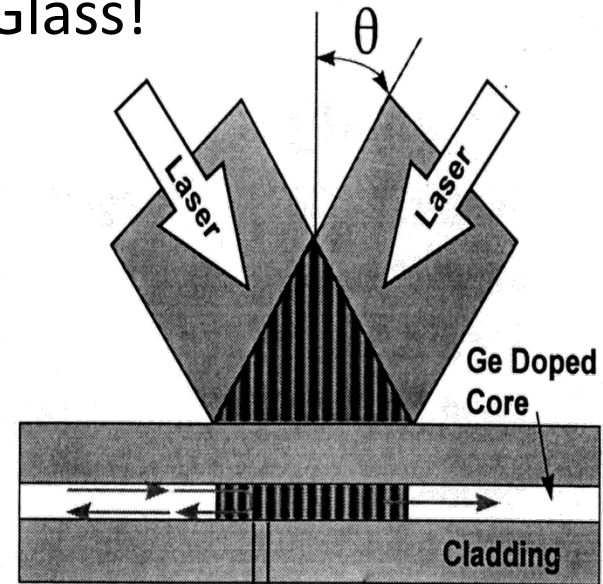
Introduction: Why Use Fiber?

- Immunity to electromagnetic interference, radio-frequency interference, and radiation.
- Compact, lightweight, ruggedized device for smart structure
 - Embedded into structure
 - Harsh environment (under water)
- The ability to be multiplexed. (100s of sensors on a single fiber).
- Ease of installation and use (single fiber vs. multitude of lead wires).
- Potential low cost as a result of high-volume telecommunications manufacturing.
- WEIGHT SAVING vs Strain gauge



Background: A Piece of Glass!

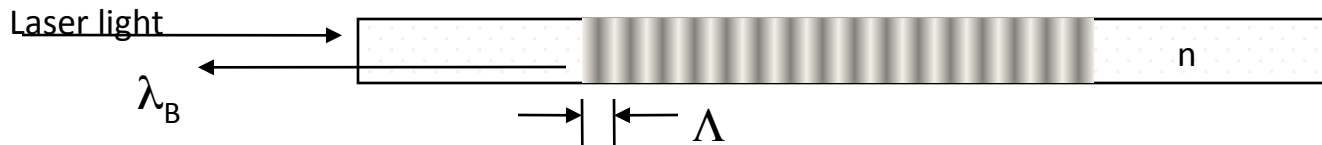
- Fiber Bragg Grating (FBG) sensor is that a change in strain state will alter the center wavelength (λ) of the light reflected from an FBG.
- A fiber's index of refraction (n) depends on the density of the dopants it contains.
- FBGs are created by redistributing dopants to create areas that contain greater or lesser amounts, using a technique called laser writing or dopant modulation.
- The index of refraction is modulated throughout the length of the grating.
- This grating reflects a narrow spectrum of light that is directly proportional to the period of the index modulation (Λ) and the effective index of refraction (n).
- The Bragg wavelength (λ_B), is expressed by $\lambda_B = 2 n \Lambda$. Because change in temperature (ΔT) and strain ($\Delta \epsilon$) directly affect Λ and n , any change in temperature or strain directly affects the λ_B .



$$\lambda_B = 2n\Lambda$$

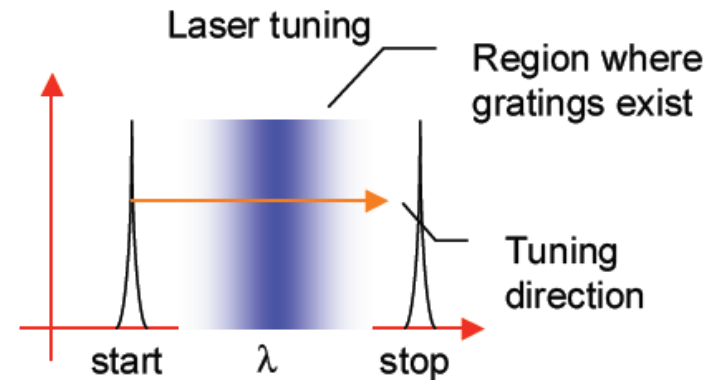
$$\Lambda = \frac{\lambda}{2\sin\theta}$$

$$\frac{\Delta\lambda_B}{\lambda_B} = K\epsilon$$



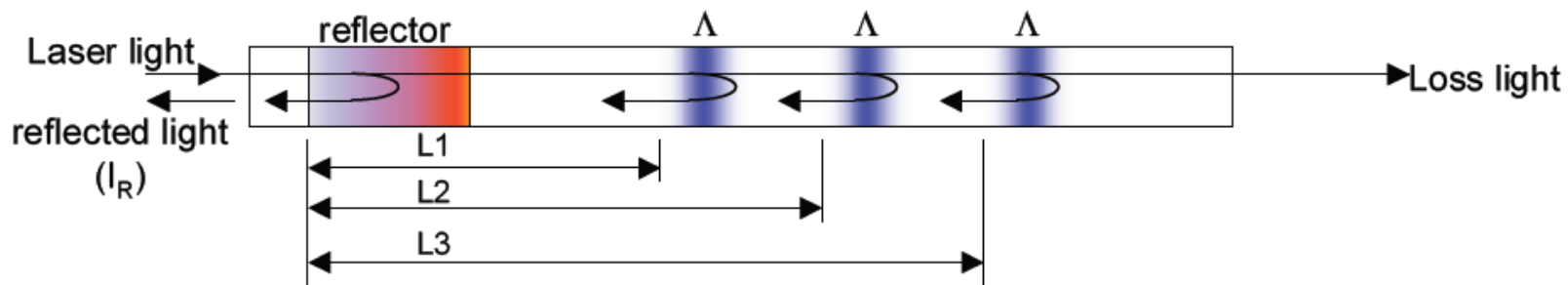
NASA Grating Modulation Multiplexing Method

- Multiplex 100s of sensors onto one fiber.
- All gratings are written at the same wavelength.
- A narrowband wavelength tunable laser source must be used to interrogate sensors.
- Sensor size can be from 0.1mm to 100mm gage lengths.

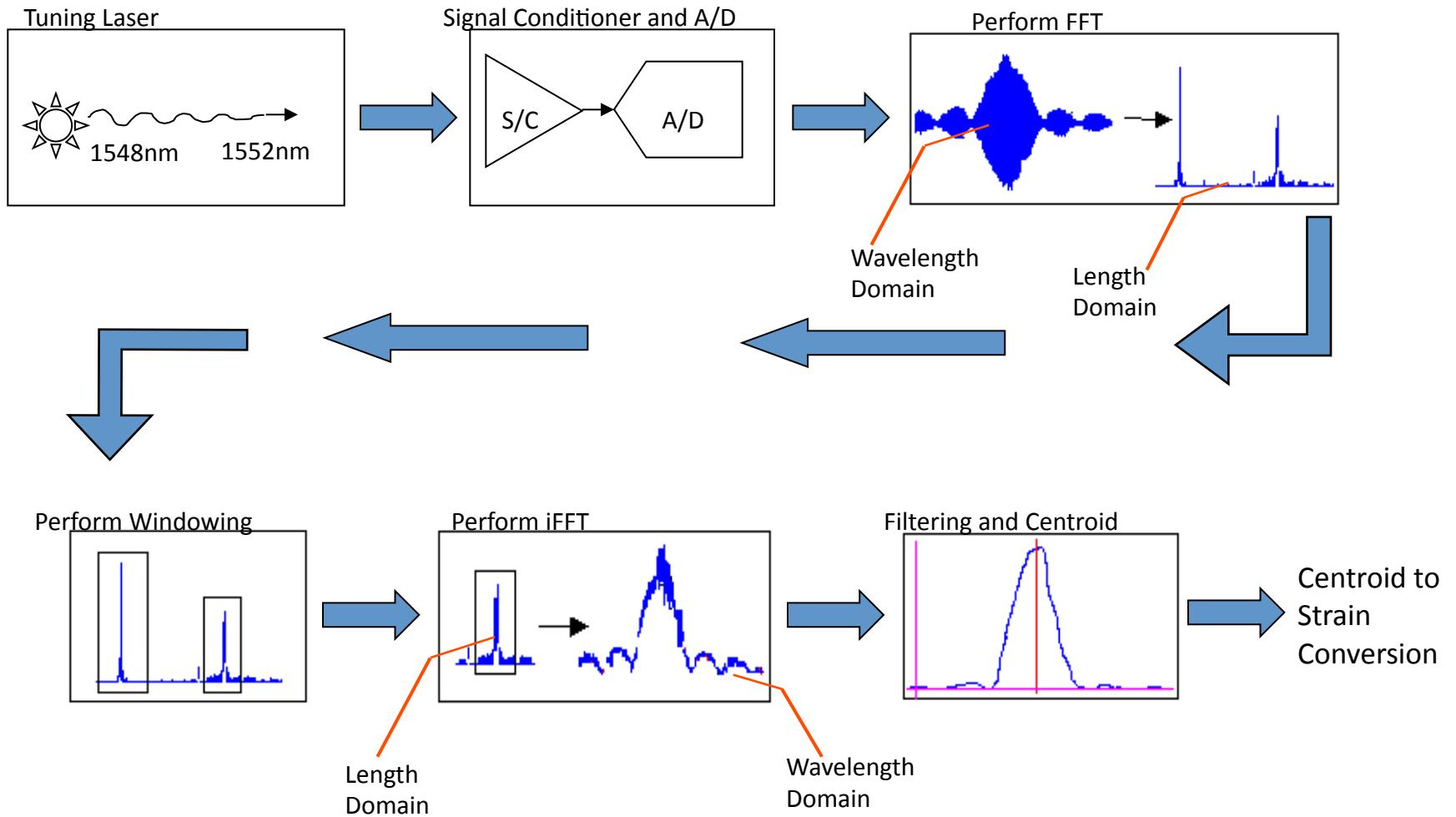


$$I_R = \sum_i R_i \cos(k2nL_i) \quad k = \frac{2\pi}{\lambda}$$

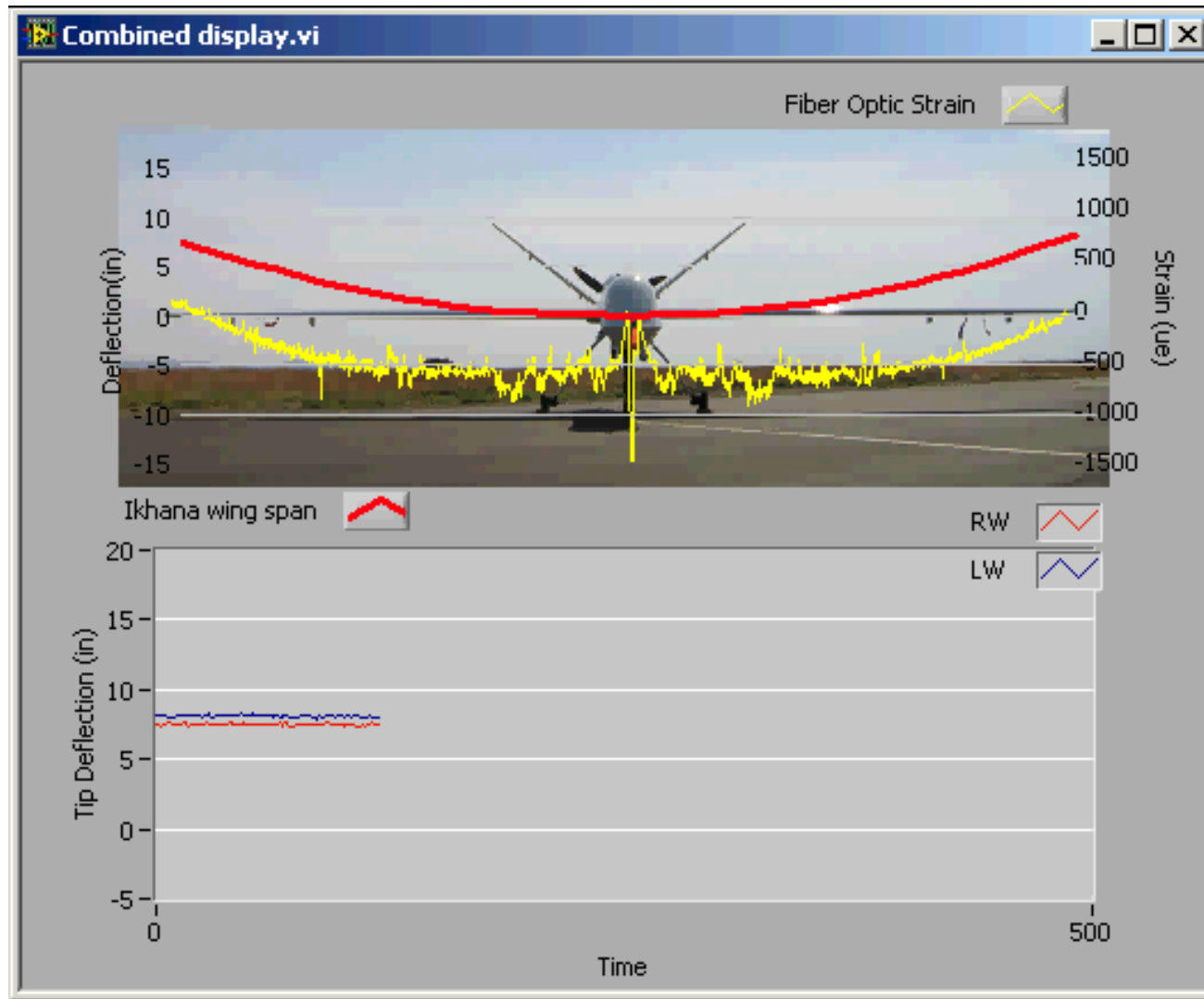
R_i – spectrum of i^{th} grating
 n – effective index
 L – path difference
 k – wavenumber



Processing Procedure



Fiber Strain Sensors in Action



National Aeronautics and Space Administration
Dryden Flight Research Center

Fiber Optics Wing Shape Sensing System (FOWSS) for Ikhana

- Fiber count: 4
- Max Fiber length: 40 ft
- Max sensing length: 20 ft
- Max gages/fiber: 480
- Total gages/system: 1920
- Sample rate: 50 Hz @ 2 fibers
30 Hz @ 4 fibers
- Power: 28Vdc @ 4 Amps
- Weight: 23 lbs
- Size: 7.5 x 13 x 13 in



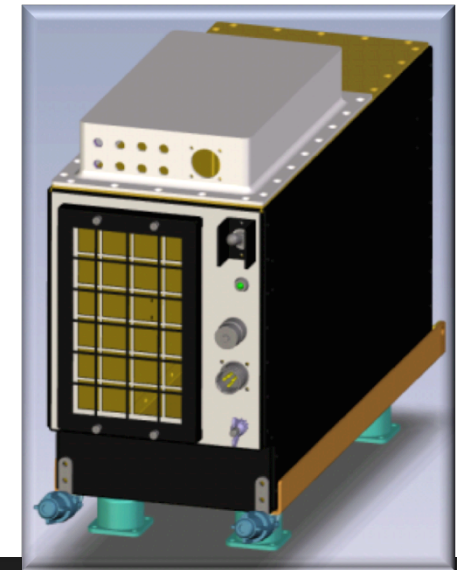
Fiber Optics Instrumentation Development System for NASA Composite Crew Module

- Fiber count 4
- Max. fiber length 40 ft
- Max sensing length 20 ft
- Max. sensors / fiber 480
- Total sensors per system 1920
- Min. grating spacing 0.5 in
- Sample rate 2 fibers @ 50 sps
4 fibers @ 24 sps
- Interface Gigabit Ethernet
- Power 120 VAC
- Weight 12 lbs
- Size 9 x 5 x 11 in



Fiber Optics Instrumentation Development System for Global Observer

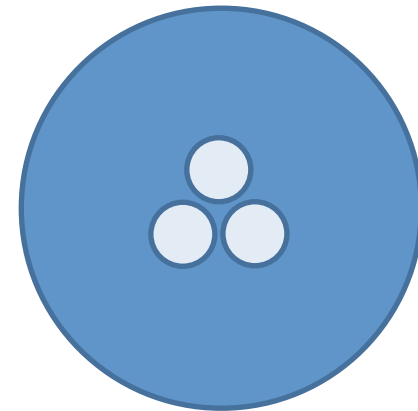
- Fiber count: 8
- Max Fiber length: 80 ft
- Max sensing length: 40 ft
- Max gages/fiber: 1000
- Total gages/system: 8000
- Sample rate: 0-50 Hz
- Power: 28Vdc
- Weight: 28 lbs
- Size: 7.5 x 13 x 18 in



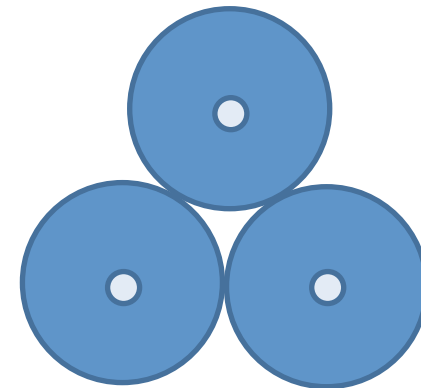
Recent Development

Shape Sensing using fiber strain sensors

- From collaboration with NASA LaRC, shape sensing using fiber strain sensors has been realized
 - Initial research focuses upon 3-core fiber
 - This speciality fiber can be replaced with 3 conventional fibers superposition from one another at 120°
- From knowing the strain value of each fiber, the 3-dimensional position of the fiber can be accurately rendered in real-time
 - Strain \rightarrow 3D Position



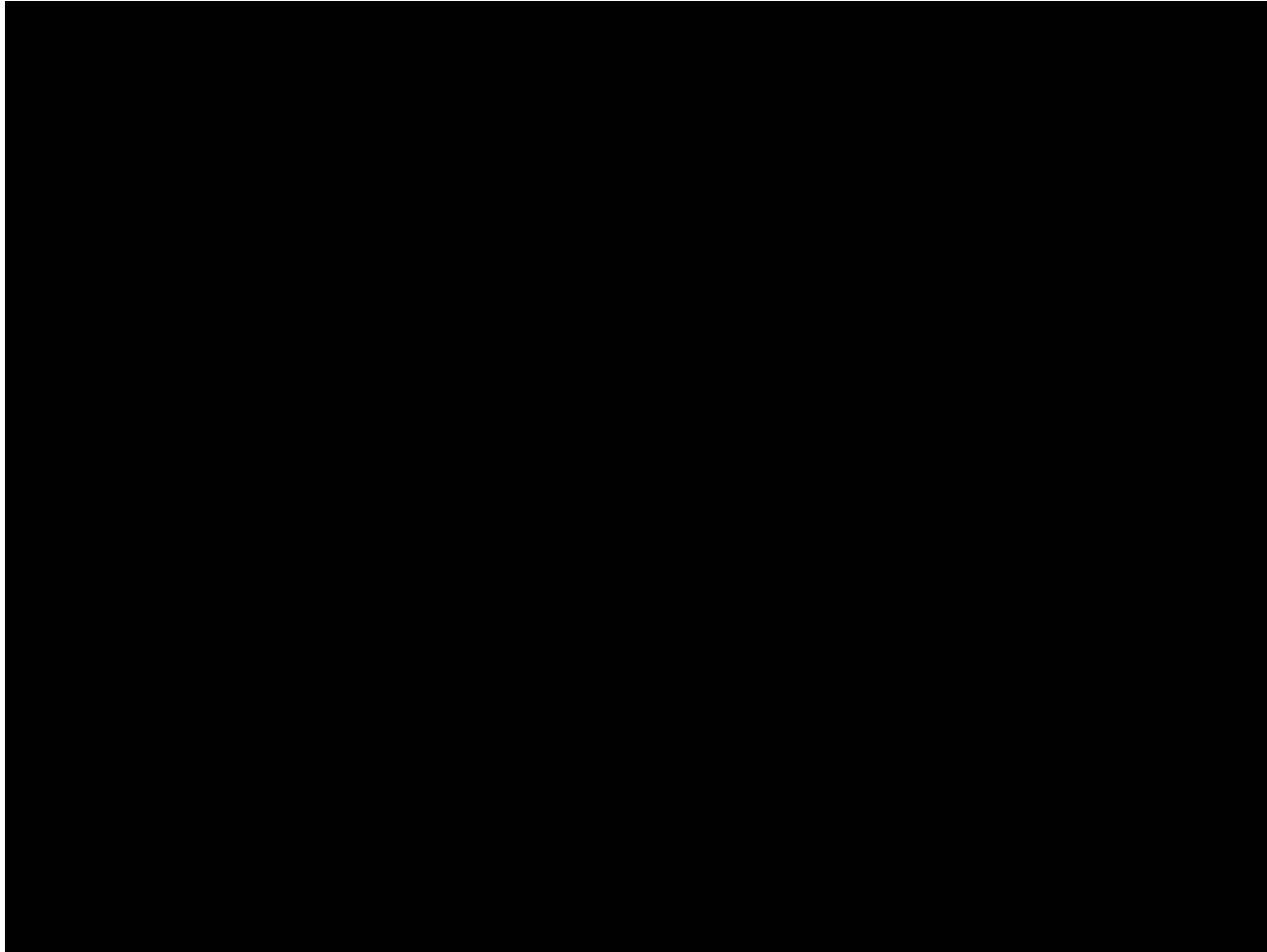
3 core fiber



3 SMFs aligned in 120°

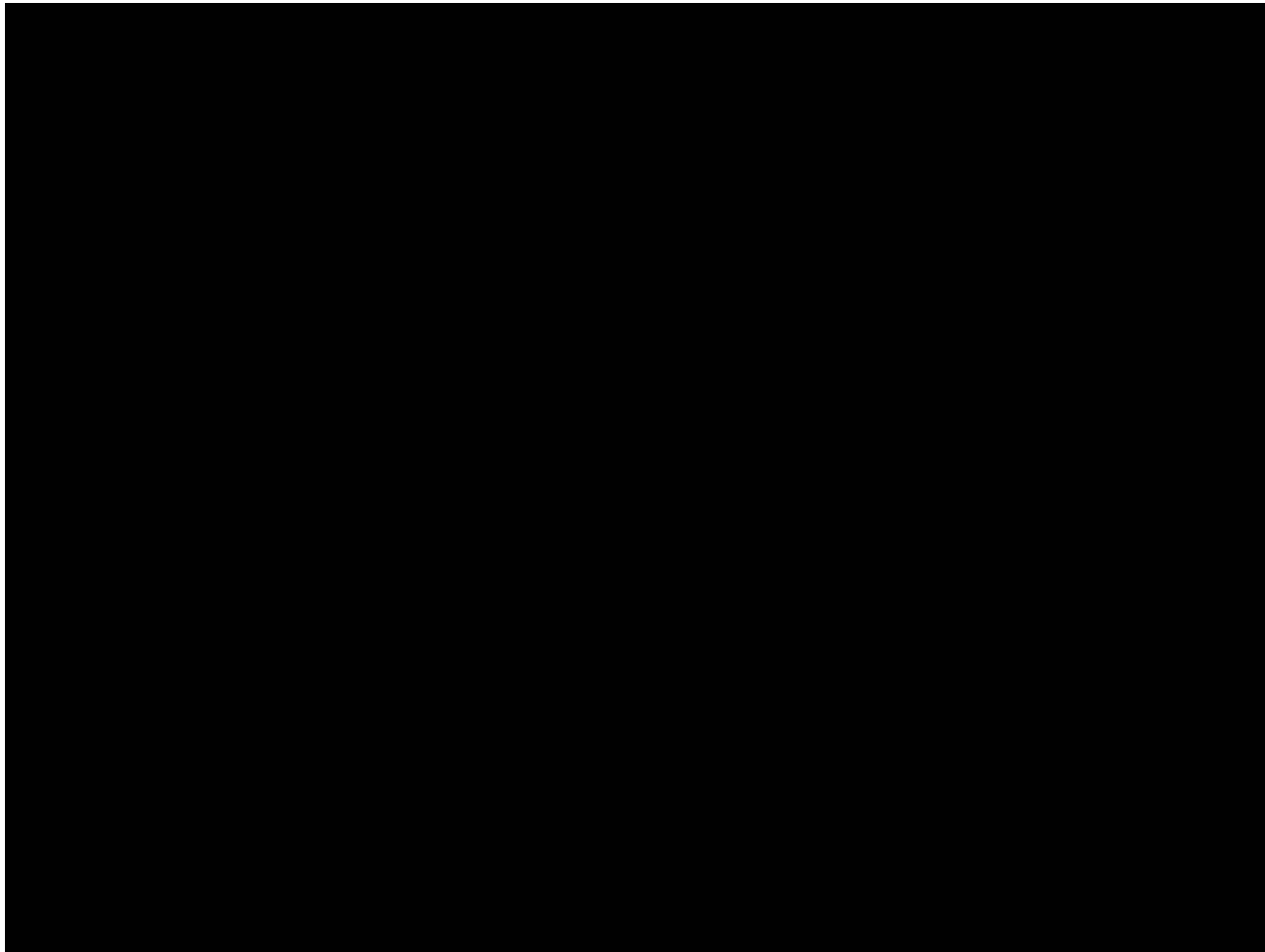


Prototype – Hexagon strain rod



National Aeronautics and Space Administration
Dryden Flight Research Center

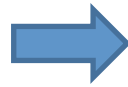
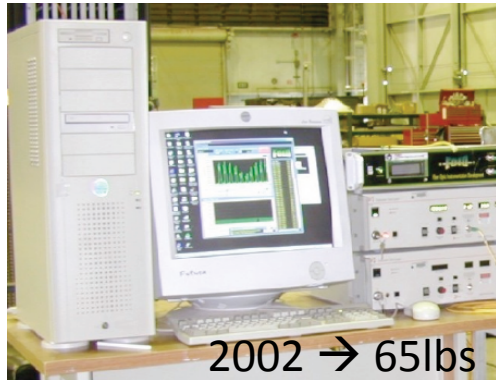
Prototype – Shape sensing fiber



National Aeronautics and Space Administration
Dryden Flight Research Center

Conclusion

- NASA DFRC has successfully develop fiber optics strain sensors technology from laboratory to real-world application



2010 and beyond
8-channel system
>100Hz
~30lbs

2-channel system
~50Hz
~10lbs

- Current status
 - Dryden FBG system are installed on Ikhana and Global Observer UAV for real time strain sensing
 - Real-time fiber shape sensing is currently being developed
- Potential application of technology beyond aeronautics
 - Automotive Sector
 - Energy Sector
 - Biomedical Sector



National Aeronautics and Space Administration
Dryden Flight Research Center