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Experimental evaluation of fatigue crack initiation from corroded hemispherical notches in aerospace structural materials

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ABSTRACT

A test program was developed and executed to evaluate the influence of corroded hemispherical notches on the fatigue crack initiation and propagation in aluminum 7075-T7351, 4340 steel, and D6AC steel. Surface enhancements such as shot peening and laser shock peening were also incorporated as part of the test effort with the intent of improving fatigue performance. In addition to the testing, fracture mechanics and endurance limit based analysis methods were evaluated to characterize the results with the objective of challenging typical assumptions used in modeling fatigue cracks from corrosion pits. The results specifically demonstrate that the aluminum and steel alloys behave differently with respect to fatigue crack initiation from hemispherical corrosion pits. The aluminum test results were bounded by the fracture mechanics and endurance limit models while exhibiting a general insensitivity to the residual stress field generated by shot peening. The steel specimens were better characterized by the endurance limit fatigue properties and did exhibit sensitivities to residual stresses from the shot peening and laser shock peening.

Keywords Aluminum 7075-T7351, 4340 Steel, D6AC Steel, fatigue, corrosion pit, initiation, nucleation, propagation, laser peen, shot peen, residual stress, threshold

NOMENCLATURE

R = load ratio (minimum load/maximum load)

S_y = yield stress

S_{ult} = ultimate stress

K_f = fatigue notch factor

K_t = stress concentration factor

A = crack depth

C = half surface crack length

σ_{th} = threshold stress

ΔK_{th} = threshold stress intensity factor range

β = geometric correction factor used in stress intensity factor calculation

P = applied load

T-S = Transverse Short material orientation

a_i = initial pit depth

$2c_i$ = initial pit surface diameter

a_f = crack depth after crack initiation

$2c_f$ = surface crack length after crack initiation

w = specimen width

t = specimen thickness

S-N = Stress versus life fatigue relationship

INTRODUCTION

Aerospace structural materials such as aluminum 7075-T7351, 4340 steel, and D6AC steel are susceptible to pitting when operated in a corrosive environment. Corrosion pits can readily act as crack initiation regions due to the metallurgical damage in addition to the stress concentration influences. The phrase crack initiation refers to the process by which a fatigue crack is formed and reaches a sufficient size to which it can then be characterized by fracture mechanics methods. Fatigue cracks emanating from naturally occurring corrosion pits are frequently modeled as surface cracks or a combination of other standard fracture models [1-7]. This project challenged that assumption and approach by utilizing idealized corrosion pits (corroded hemispherical notches) that are subsequently tested to observe the stress concentration and corrosion effects on the fatigue crack initiation and propagation behavior.

Current fracture mechanics tools like NASGRO [8] and AFGROW [9] do not currently have the capability to capture the geometry of cracks emanating from hemispherical notches. The desire to enhance the fracture mechanics understanding of crack initiation and propagation from an idealized corrosion pit motivated this project. The overall objective was to advance the fundamental understanding of the mechanics of fatigue crack initiation and propagation processes from idealized corrosion pits in Al 7075-T7351, 4340 steel, and D6AC steel with a variety of surface enhancements that include shot peening or laser shock peening. The approach consisted of performing fatigue crack initiation and propagation testing that encompassed the three alloys, surface treatment conditions, multiple corrosion pit sizes, and two applied load ratios. The results were then studied to evaluate the applicability of using surface crack approximations or endurance limit approximations to characterize the initiation of fatigue cracks from idealized corrosion pits.

MATERIALS AND PROPERTIES

Three aerospace materials were chosen for the testing program including aluminum 7075-T7351, 4340 steel, and D6AC steel. All three materials were procured as plate (Al 7075) or forging (Steels) and the specimens were tested in the transverse-short (T-S) orientation with the various surface treatment conditions. The yield stress, σ_y , ultimate stress, σ_{ult} , and specific surface enhancements are listed in Table 1 for reference.

Surface Enhancements

Previous efforts have examined the effects of surface enhancements on the fatigue performance of aluminum and steel alloys and revealed noticeable improvements in the resistance to fatigue crack initiation [10-12]. For this reason, shot peening and laser shock peening surface enhancements were used in this program so that their effects on the corroded notches could be evaluated. The specific surface enhancement parameters for each surface condition are provided in Table 1. From each set of specimens selected for fatigue testing, one was chosen specifically for residual stress measurements by using the slitting method [13,14]. This technique is destructive, so the selected specimens could not be used for further fatigue testing. For reference, none of the bare specimens from any of the three materials were selected for residual stress measurements.

Surface treatments on the aluminum 7075-T7351 consisted of a nominal shot peen and a heavy shot peen (or heavy peen), and the results from these residual stress measurements are shown in Fig. 1a along with

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3 the respective notch depths identified for reference. A graphical schematic of the sectioned corrosion pit
4 geometry with respect to the residual stress field is also shown in Fig. 1a. It is apparent that the heavy
5 peen specimen has a larger magnitude peak compressive stress but a steeper stress gradient compared to
6 the nominal shot peen condition. For both cases, the residual stress field becomes tensile at
7 approximately 1.27 mm into the depth of the specimen. With respect to the corrosion pit dimensions, the
8 smaller pits in the nominal shot peened specimens were completely encompassed by the compressive
9 residual stress field, but the root depths of the larger pits crossed into the tensile region. The root depth
10 for all of the pits in the heavy peened specimens crossed into the tensile region. This implies that the
11 effect of the residual stress was more pronounced at the surfaces of the corrosion pits rather than the root
12 of the pits.
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17 The 4340 steel specimens had either the laser shock peening (or laser peening) surface enhancement
18 method or the shot peening, and the residual stress measurements for both cases are shown in Fig. 1b. An
19 inset is shown within Fig. 1b to examine the compressive region in greater detail and identify the depth of
20 the corrosion pits. Although the surface residual stress for the two methods was approximately the same
21 ($\sim -800\text{MPa}$), the laser peened residual stress remained compressive up to a depth of approximately 2.03
22 mm while the shot peening residual stress became positive at approximately 0.51 mm. These results
23 demonstrate that the laser peening residual stress remained compressive over a distance of four times that
24 of the shot peening surface. It should also be noted that the laser peen resultant peak tensile stress is also
25 approximately four times greater than the peak shot peened resultant peak tensile stress. The depth of the
26 notch root for the smallest pit tested in the shot peened condition corresponds with zero residual stress,
27 but the larger pit sizes had depths that spanned into the residual tensile stress field. All of the laser
28 peened specimens had corrosion pits that were encompassed by the compressive residual stress.
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33 The D6AC steel specimens were only tested in the bare or shot peened conditions, and the results of the
34 shot peening residual stress measurements are shown in Fig. 1c with the pit depth ranges identified. It is
35 apparent that root of the smallest pit depth crosses into the tensile residual stress region. All shot peened
36 D6AC steel specimens had corrosion pits in which some region of the pit crossed into the tensile residual
37 stress field.
38

39 **Notch Fatigue Properties**

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41 The fatigue and fracture characteristics of cracks emanating from idealized corrosion pits was the primary
42 motivation of this project, so in order to fully explore these aspects, the influence of the hemispherical
43 notches needs to be discussed. A hemispherical notch should not be confused with a semi-circular notch
44 because the boundary conditions are different. A hemispherical notch is a 3-dimensional feature and
45 cannot be generalized in 2-dimensions like a hole in a plate because the added dimensional constraint in
46 three dimensions essentially lowers the stress concentration. The analytical stress concentration factor for
47 a hole in an infinite plate is $K_t=3.0$, whereas, the stress concentration factor, K_t , for a hemispherical notch
48 is reported as $K_t=2.15$ in [15] and $K_t=2.23$ in [16]. For the remainder of this document, the stress
49 concentrator at a hemispherical notch is assumed to be $K_t = 2.23$.
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54 In the mechanics of fatigue, materials are sensitive to the stress concentrator in a variety of ways. The
55 fatigue notch factor, K_f , represents the reduction in remotely applied stress required to initiate a crack
56 from a notch. The notch sensitivity, q is shown in equation 1 to be a function of the stress concentrator,
57 K_t and the fatigue notch factor K_f [17,18]. The value for q can also be determined empirically, where α is
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the notch sensitivity parameter for a given material and ρ is the radius of the notch. For the materials used in this study, α is 0.51 mm for aluminum, 0.06 mm for 4340 steel and 0.08 mm for D6AC steel[17]. These values are assumed to be valid for hemispherical notches although they were developed from planar stress concentrators.

$$q = \frac{K_f - 1}{K_t - 1} = \frac{1}{1 + \frac{\alpha}{\rho}} \quad (1)$$

It can be seen in equation 1 that as the notch sensitivity, q , increases and approaches unity (1.0), the notch fatigue factor, K_f , approaches the value of the stress concentration factor, K_t . This means that larger pit sizes have greater influence on the fatigue life. As the notch sensitivity, q , approaches zero, the notch fatigue factor, K_f , approaches unity. This means that the effect of the pit on fatigue life becomes negligible for small pit sizes irrespective of material.

For reference, the fatigue notch factor, K_f , is shown as a function of pit depth in Fig. 2 and demonstrates that the aluminum alloys are much less sensitive to notches than the steel alloys for a variety of pit depths. For example, a hemispherical notch between 0.51mm and 1.52 mm in aluminum will have a fatigue notch factor between 40-85% of the stress concentration factor, but the same pit size in steels will have a notch fatigue factor that is over 90% of the stress concentrator.

EXPERIMENTAL METHODS AND PROCEDURES

Specimen Design and Preparation

The general approach for developing the fatigue test was to incorporate a dogbone shaped specimen design where the grip section is wide and tapered to the gage section so that the highest stresses are in the gage section. Within this specimen, the hemispherical notch was centrally located. For each material used, the nominal gage thickness varied between 10.2-12.7mm and the width was approximately 22.9mm. A schematic of a representative dog-bone specimen is provided in Fig. 3. Once the machining process was complete, a set of specimens from each material were then either shot peened or laser shock peened. After the surface treatments were conducted, the specimens were then prepared for the notching and corrosion pit processes.

The notching and pitting processes for the specimens included the following steps with the specific nuances for each material highlighted. All specimens were polished, primed, and then coated with an acrylic paint to protect the material not intended for corrosion. A small hemispherical notch was then centrally located and drilled into the specimen as shown in Fig. 3. For the aluminum specimens, a corrosion cell was created by filling a polyvinyl chloride tube attached to the specimen with a mixture of 240 mL de-ionized water, 1.6 mL HNO₃ (nitric acid), and 0.8 g NaCl. The corrosion process was allowed to react from 10 - 60 minutes depending on the resulting pit size. The steel specimens were corroded in the hemispherical notches by applying four drops of 60% de-ionized water and 40% concentrated nitric acid and maintaining the reaction for approximately 45 minutes. After the pitting process, the corrosion cell was removed (for the aluminum specimens) and the specimens were rinsed with water. The coating used to protect the base material was then removed with a solvent.

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3 At this stage, the specimens were measured and instrumented for direct current potential drop for the
4 Fracture Technology Associates (FTA) data acquisition system. The potential drop method could detect
5 the crack propagation from the notch through the change in electric potential due to the change in cross
6 sectional area. After the specimens were prepared for data acquisition, they were then tested.
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9 **Fatigue Test Methods**

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11 The process for performing the corrosion pit initiation and propagation tests is depicted in Fig. 4 and
12 discussed here. All testing was performed in servo-hydraulic test machines. During the beginning of the
13 test phase, the specimens were cycled at a given starter load and load ratio, $R=0.1$ or $R=0.7$, until a crack
14 was detected or 3 million cycles was reached. Each test machine was operated at the maximum capable
15 frequency to ensure that the loads were reached and that the system avoided natural frequencies. If the
16 FTA system detected a crack of approximately 0.254mm, the crack was visually confirmed and the test
17 halted or the test was continued until a visual crack was detected and confirmed. If 3E6 cycles was
18 reached prior to detecting a crack, the load was then increased approximately 10% and the test was started
19 again until a crack was detected or 3E6 cycles was reached. This process is known as step testing and is
20 used to determine a threshold load or endurance limit depending on the critical parameter. The process
21 was repeated until a crack was detected in the pit or anywhere else on the specimen.
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26 The initial growth is referred to as the initiation phase for this study. Because the cracks did not grow
27 uniformly out of the idealized corrosion pits, the initial growth was measured and photo-documented after
28 the test. In order to measure the initial growth, the fracture surface was marked after the initiation phase
29 of the test by either changing the load ratio from $R=0.7$ to $R=0.1$ or increasing the load if the load ratio
30 during the initial test was $R=0.1$. Once the crack marking occurred, the FTA was programmed to then
31 stop the test after the crack reached 3.81mm in depth. At this point, the specimen was fractured to expose
32 the crack face and subsequently, the pit dimensions, initial growth, and final growth measurements were
33 documented. A total of 139 specimens were tested during this project and the general test matrix,
34 provided in Table 1, includes the material, surface condition tested, load ratio, and number of tests.
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38 **FATIGUE TEST RESULTS**

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40 The results from the fatigue step testing provided information regarding the threshold or initiation
41 properties of crack emanating from idealized corrosion pits. The terms threshold and initiation are used
42 in this context because it is not clear if the initial crack process is dominated by fracture mechanics
43 (threshold) or fatigue (initiation). One advantage of this test approach is that the threshold behavior can
44 be determined rather rapidly if the initial loads are appropriately chosen. The disadvantage is that the
45 initial predictions may be too high resulting in a first step initiation, or too low resulting in several steps
46 and cycles required to reach threshold. The test results for all three materials are provided in Table 2
47 through Table 9 and shown in Fig 12 through Fig 14. Within these tables are measurements, all load
48 steps, cycles, and crack length measurements. The initial crack measurement, a_i and $2c_i$, indicate the pit
49 depth and surface width respectively. The crack measurements after the initiation process are represented
50 by a_f and $2c_f$. For simplicity, the test results are organized in bins that correspond to a range of small,
51 medium, and large pit depths. For example, a bin of 0.254mm indicates all pit depths tested between
52 0.254 mm and the next bin size (i.e. 0.762 mm).
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57 **Aluminum 7075-T7351 Initiation Test Results**

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3 The test results for each of the different surface conditions and load ratios are included in Table 2 through
4 Table 4 for the aluminum specimens. Several of the initial tests were used to understand the process, so
5 these tests failed in the spot weld or exhibited some other problem during testing. In general, the primary
6 nuances experienced during these tests included first step failures and a few spot weld crack initiations.
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9 The results from the aluminum tests are shown in Fig. 5 as maximum stress as a function of cycles to
10 crack initiation. The runout data are not included in the plots, but the data are provided in the tables. The
11 data are also shown with the S-N curve fit of a similar alloy, Al 7075-T6, with fatigue notch factor, of
12 $K_f=1.8$ which is in the same range as the test data ($K_f=1.4-1.9$) [19]. In addition to the fatigue curve, the
13 threshold stress corresponding to a semicircular surface crack based on an initial flaw size and threshold
14 stress intensity factor range is also shown for reference. The threshold stress, σ_{th} is determined by using
15 equation 2 where ΔK_{th} is the threshold stress intensity factor range, R is the load ratio, β is the
16 dimensionless shape factor obtained from NASGRO [8], and a is the semi-circular crack radius.
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$$\sigma_{th} = \frac{\Delta K_{th}}{(1 - R)\beta\sqrt{\pi a}} \quad (2)$$

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24 Upon initial observation, all of the $R=0.1$ nominal shot peen and heavy peen test results are conservative
25 compared to the S-N fatigue curve. This includes both the shot peen and heavy peen test results. In
26 addition, the threshold stress based on a surface crack fracture model is a conservative estimate compared
27 to all test results. These results imply that the $R=0.1$ test results are essentially bounded by the threshold
28 stress and fatigue curve. It is difficult to discern the effect of shot peening on these test results, but this
29 issue will be discussed further in the threshold assessment section.
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33 The $R=0.7$ shot peen test results appear to be well characterized by the fatigue curve for both shot peening
34 cases. This initial result suggests that shot peening does not influence the initiation life of corrosion pits
35 for 7075-T7351 and that the propagation life is negligible compared to the initiation life. The fatigue
36 curve appears to be conservative compared to all of the $R=0.7$ data in the bare condition, but there are a
37 few shot peened and heavy peened tests in which the fatigue curve is non-conservative. Again, the
38 threshold stress for a surface flaw is very conservative compared to the $R=0.7$ test results, but does
39 provide a lower bound estimate of the results.
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42 **4340 Steel Initiation Test Results**

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44 The 4340 steel test results for each of the bare, shot peened, and laser peened specimens are included in
45 Table 5 through Table 7 and shown in Fig. 6. Several of the laser peened tests proved difficult to nucleate
46 a crack within the pit. This result alone should be noted because the pit in steel had a notch fatigue factor
47 of $K_f=2.1-2.18$ that is very similar to the stress concentration factor of $K_t=2.2$. This implies that the
48 stress concentration effect is not drastically reduced in the steel alloys under cyclic loading, but there
49 were still problems nucleating a crack in the pit.
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53 All test nuances are documented but the most common problems included first step cracking, spot weld
54 cracking, and cracking away from the pit. A total of seven specimens initiated cracks either at the tack
55 weld or away from the corrosion pit and five of these specimens had pit sizes that were smaller than 0.76
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3 mm. Only two specimens out of five with laser peening and a load ratio of $R=0.1$ initiated cracks in the
4 pits.
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7 For comparison, the fatigue curve from the 4340 steel with a fatigue notch factor of $K_f=2.9$ is shown in
8 Fig. 6 [19]. This notch fatigue factor is more conservative than the test data which had fatigue notch
9 factors in the range of $K_f=2.1-2.8$. Observation of the test results demonstrate that the fatigue curve was
10 conservative compared to all the test data for both load ratios. This result is expected because of the
11 higher fatigue notch factor in the fatigue curve compared to the corrosion pit specimens. The figures also
12 demonstrate the threshold stress corresponding to a semicircular surface crack based on an initial flaw
13 size and threshold stress intensity factor range calculated from equation 2. Again, the surface crack based
14 threshold appears to be conservative compared to the test results.
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18 Upon initial observation, it is apparent that the shot peened and laser peened specimens provide benefit
19 toward the fatigue initiation stress. This is very evident with the $R=0.1$ test results. The laser peen
20 specimens exhibit the most improvement followed by the shot peened specimens. None of these results
21 have been post processed to account for residual stresses, but it is believed that the benefit is due to the
22 residual stress effects from the surface enhancements.
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24 **D6AC Steel Initiation Test Results**

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26 The test results for the bare and shot peened D6AC with corresponding load ratios are included in Table 8
27 and Table 9 and shown in Fig. 7. Within Fig. 7, the S-N data of D6AC steel with a fatigue notch factor of
28 $K_f=2.0$ is also shown for reference. This curve is non-conservative compared to the test data which has a
29 fatigue notch factor of $K_f=2.09-2.16$ [20]. Two caveats regarding the S-N data that should be discussed
30 include the strength of the S-N material is not the same as the corrosion pit material and the load ratios are
31 similar but not the same ($R=0.0$ vs. $R=0.1$ and $R=0.5$ vs. $R=0.7$) [20]. The tensile strength of the material
32 associated with the S-N data is 1861MPa whereas the corrosion pit material strength is 1103MPa. It is
33 not clear how the strength influences the fatigue data for this material, but the most dramatic effects
34 would occur at stresses near or above 551MPa, which is the yield stress for the D6AC tested in this
35 program. The figures also show the threshold stress corresponding to a semicircular surface crack based
36 on an initial flaw size and threshold stress intensity factor range. Again, the surface crack based threshold
37 appears to be overly conservative compared to the test results.
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43 Upon initial observation, it is apparent that the shot peening provides some fatigue benefit for the $R=0.1$
44 tests with the larger pit diameters. The initiation stresses of the smaller pit diameter specimens do not
45 exhibit the same improvement and correspond with the bare test results. It is difficult to discern any
46 benefit in the $R=0.7$ test results for all pit sizes.
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48 **FATIGUE THRESHOLD ASSESSMENT**

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50 The results shown in the previous section provide a method of assessing the applied stress as a function of
51 cycles to determine the threshold. One of the disadvantages associated with this method of displaying the
52 results is that the initiation cycles from a previous step are not included. A different approach to
53 displaying these results includes plotting the applied stress versus the pit depth, but this approach does not
54 account for the load associated with the previous runout step. This section discusses an approach to
55 utilize the test results to establish the threshold stress.
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The step test approach used to determine threshold implies that the true initiation and/or propagation threshold occurs somewhere between the previous load step and the current load step. In order to account for this, an equation can be used that interpolates the threshold load based on the applied loads and applied cycles [21]. The interpolation equation is given below in equation 3 where P_{th} is the threshold load, P_{ps} is the previous load from the previous step, P_f is the current load step, N_f is the number of cycles in the current step, and N_{step} is the number of cycles per step. The threshold load can then easily be converted to stress and then threshold stress can be shown as a function of pit depth. All of these parameters are available in Table 2 through Table 9 for the test results.

$$P_{th} = P_{ps} + \frac{(P_f - P_{ps})N_f}{N_{step}} \quad (3)$$

Another advantage of the described approach is that these results can be compared to an analytic threshold prediction based on either the endurance limit or fracture mechanics. The fracture mechanics approach uses equation 2 previously discussed where the crack length is set equal to the notch radius and the threshold SIF corresponds to the specific material properties. For a variety of notch depths, the trend can be established and compared to the test data. The other method used in assessing the threshold stress was to use the S-N endurance limit and adjust it to account for the notch. For most cases, the fatigue curve from literature was used to estimate the endurance limit associated with 2E6 cycles using the appropriate load ratio of either $R=0.1$ or $R=0.7$ [19,20]. This endurance stress was then reduced to account for the fatigue notch factor, K_f , and respective pit diameter from equation 1. This is not the same as using notched fatigue properties. It is a method of adjusting the unnotched parameters to assess the effect of a notch.

Aluminum 7075-T7351 Threshold Assessment

The test results showing threshold stress as a function of pit depth are displayed in Fig. 8. In addition to the test results, the surface crack threshold and endurance limit predictions are also shown. The results demonstrate that an endurance limit prediction is non-conservative for the aluminum and the fracture mechanics approach is overly conservative. Closer observation of the test results from both load ratios reveals that the aluminum pit initiation appears to be insensitive to the residual stress. This means that the test results do not appear to be influenced by the surface treatments for the range of pit depths from 0.3mm – 2.0 mm. This is demonstrated by the fact that there does not appear to be an increase in initiation stress for the shot peened specimens. This observation is only valid for the pit sizes tested and may show greater influence at smaller pit sizes.

4340 Steel Threshold Assessment

The 4340 steel test results are shown in Fig. 9 as threshold stress as a function of pit depth. In addition to the test results, the surface crack threshold predictions are shown with the endurance limit prediction. Initial observations reveal that the surface crack approximation is overly conservative. Closer examination reveals that the unpeened threshold stress response for $R=0.1$ appears to be constant with respect to the pit diameter and this implies that the initiation process may be controlled by fatigue (i.e. endurance limit) rather than fracture mechanics. This observation is also supported by the fact that the notch fatigue factor, K_f , for steel alloys does not vary much for pit sizes above 0.51mm. The shot-peened

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3 and laser peened specimens do not follow the fracture mechanics predictions either, but the curve does
4 not account for residual stress. The effect of residual stress is apparent for the $R=0.1$ threshold results
5 with laser peening resulting in the most increase in threshold stress.
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8 The $R=0.7$ shot peening and laser peened specimens still show some benefit, but it is not as pronounced
9 as in the $R=0.1$ test results. This may be explained by the fact that the localized stresses are above yield,
10 so the influence of shot peening or laser shock peening may not be as apparent as they are in the $R=0.1$
11 results. Although the endurance limit prediction for the $R=0.7$ case is conservative, the surface crack
12 prediction is overly conservative.
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14 **D6AC Steel Threshold Assessment**

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17 The threshold stress predictions for the D6AC specimens are shown in Fig. 10 as threshold stress as a
18 function of pit depth. It is immediately apparent that the surface crack predictions are overly conservative
19 compared to the test data for both load ratios. The endurance limit predictions for $R=0.1$ characterize the
20 larger pits ($>1.02\text{mm}$) better than the smaller pits. It should be noted that the $R=0$ endurance limit for
21 1861MPa D6AC steel was used for the prediction[20]. The $R=0.1$ endurance data would most likely fit
22 better, but it is difficult to discern what the effect of reduced strength would have on the endurance limit.
23 For the $R=0.1$ test results, the residual stress from the shot peening did appear to have the effect of
24 increasing the threshold stress at the larger pit diameters.
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28 The $R=0.7$ endurance limit predictions are very conservative compared to the test results. These results
29 are for an $R=0.5$ load ratio instead of $R=0.7$, so the actual discrepancy may not be as significant if the
30 predictions were based on $R=0.7$ data. The tests do appear to show an insensitivity to pit diameter at the
31 higher load ratio. This would imply that the D6AC steel can be better characterized with the endurance
32 limit rather than the fracture mechanics threshold approach. It is also interesting to note that the high load
33 ratio test data was performed well above the yield stress. Again, the residual stress effects for the $R=0.7$
34 test results were not obvious, so the effect of localized yielding may have had an effect on these
35 specimens.
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38 **CONCLUSIONS**

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40 In summary, a test program was developed and executed to evaluate the influence of hemispherical
41 corrosion pits on the initiation and propagation in aluminum and steel alloys. In addition to the general
42 corrosion pit influences, the effects of surface enhancements such as shot peening and laser shock peening
43 were also evaluated. The specific conclusions for the test program and analysis are provided below:
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- 46 1. Aluminum 7075-T7351 fatigue initiation from a hemispherical corrosion pit appears to be
47 insensitive with respect to nominal and/or heavy shot peen surface enhancements. This
48 observation is independent of corrosion pit size (pit depth= 0.25mm to 2.03mm) and applied load
49 ratio ($R=0.1$ or $R=0.7$).
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- 51 2. The fatigue initiation from a hemispherical corrosion pit in 4340 steel is sensitive to both shot
52 peening and laser shock peening surface enhancements. The shot peening provides benefit over
53 the bare condition and the laser peening provides benefit over both the bare and shot peening
54 conditions. This observation is more pronounced at a load ratio of $R=0.1$ with pit depths ranging
55 from 0.51mm - 1.02mm .
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3. D6AC steel fatigue initiation from a hemispherical corrosion pit appears to be sensitive to shot peening at a load ratio of $R=0.1$, but insensitive to shot peening at $R=0.7$. This observation includes all pit depths tested from 0.51-1.52 mm.
4. A threshold stress prediction was implemented using typical fracture mechanics assumptions to capture the step test approach and predict the applied stresses at which surface cracks will initiate from corrosion pits. This prediction method is much too conservative for the three materials tested to capture the real behavior of a crack initiating and propagating out of a corroded hemispherical notch.
5. An endurance limit based prediction was also implemented that accounts for the fatigue notch factor. This approach was overly conservative for the aluminum tests, but provided a reasonably conservative estimate for the steels. This was particularly true for the 4340 steel.

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Table 1 Material properties and surface enhancement method.

Table 2 Test results for bare aluminum 7075-T7351 corrosion pit specimens.

Table 3 Test results for shot peened aluminum 7075-T7351 corrosion pit specimens.

Table 4 Test results for heavy shot peened aluminum 7075-T7351 corrosion pit specimens.

Table 5 Test results for bare 4340 steel corrosion pit specimens.

Table 6 Test results for shot peened 4340 steel corrosion pit specimens.

Table 7 Test results for laser peened 4340 steel corrosion pit specimens.

Table 8 Test results for bare D6AC steel corrosion pit specimens.

Table 9 Test results for shot peened D6AC steel corrosion pit specimens.

Fig. 1 Residual stress measurements for the aluminum 7075-T7351 (a), 4340 steel (b), and D6AC steel (c).

Fig. 2 Fatigue notch factors as a function of pit radius for aluminum and steel alloys.

Fig. 3 Specimen schematic and reference corrosion pit location.

Fig. 4 Fatigue crack initiation and propagation step test process.

Fig. 5 Al 7075-T7351 test results for bare and nominal shot peened specimens, R=0.1.

Fig. 6 4340 steel fatigue test results for all surface conditions, R=0.1.

Fig. 7 D6AC steel bare and shot peened test results for R=0.1.

Fig. 8 Threshold stress as a function of pit depth for aluminum 7075-T7351.

Fig. 9 Threshold stress for the 4340 steel as a function of pit depth.

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3 Fig. 10 Threshold stress for the D6AC steel as a function of pit depth.
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Table 1 Material properties and surface enhancement method.

Material/ Orientation	σ_y MPa	σ_{ult} MPa	Surface Condition Tested	Surface Enhancement Parameters	# of Specimens	
					R=0.1	R=0.7
Aluminum 7075-T7351 (T-S)	248	503	Bare	NA	14	12
			Nominal Shot Peen	8-12 Almen, 100% coverage, 2 times	11	4
			Heavy Peen	5-7 Y2 Almen, 100% coverage, 5 times	12	10
4340 Steel (T-S)	1206	1310	Bare	NA	9	8
			Shot Peen	8-12 Almen 100% coverage	9	5
			Laser Peen	power density 8GW/cm ² , 18ns pulse, 16J laser energy, 3 layers, 3.33mm x 3.33mm spot size	5	6
D6AC Steel (T-S)	551	1103	Bare	NA	8	8
			Shot Peen	8-12 Almen, 100% coverage	9	9

Table 2 Test results for bare aluminum 7075-T7351 corrosion pit specimens.

w mm	t mm	load steps kN	Cycles /step E6	final P_{max} kN	final S_{max} MPa	final step cycles	a_i mm	$2c_i$ mm	a_f mm	$2c_f$ mm
Bare Aluminum 7075-T7351 R=0.1										
22.9	12.2	NA	NA	NA	NA	NA**	0.28	0.84	NA	NA
22.9	12.2	NA	NA	NA	NA	NA**	0.28	0.84	NA	NA
22.9	12.2	38,43,48,53	2	53.5	195.5	129310	0.30	0.79	NA	NA
22.9	12.2	50	NA	50.2	183.5	127822*	0.28	0.76	NA	NA
23.1	12.4	50	NA	49.2	176.6	485498*	0.33	0.84	NA	NA
22.9	12.4	30,32,33,35	3	35.7	129.6	34936	0.76	1.65	NA	NA
22.9	12.4	33	3	33.4	119.5	470751*	0.71	1.37	1.04	2.03
23.1	11.7	31	NA	31.1	110.8	186126*	0.71	1.80	1.24	2.74
22.9	12.4	18,22,27,31,36	3	35.6	128.6	172309	1.24	2.54	1.70	3.86
22.9	12.4	22,27,31	3	31.1	112.0	395825	1.22	2.49	2.08	4.72
22.9	12.4	27,31,36,40	3	40.0	144.3	1373845	0.81	2.01	1.19	2.82
22.9	12.4	22,27,31	3	31.1	112.7	361893	1.24	2.54	1.85	4.04
23.1	12.4	27	NA	26.7	95.4	272356*	0.76	1.85	1.40	2.97
22.9	12.4	27,36	3	35.6	128.3	228735	1.27	2.95	1.63	3.45
Bare Aluminum 7075-T7351 R=0.7										
22.9	12.4	64,70,77,84,	3	84.5	304.4	229299	0.71	1.50	NA	NA
22.9	12.4	80	NA	80.1	290.2	303069*	0.66	1.47	0.86	1.78
22.9	12.4	69	NA	73.4	260.2	140209*	0.76	1.88	1.52	3.12
23.1	12.4	70	NA	70.1	251.1	217381*	0.79	2.29	1.35	3.18
23.1	12.4	72	NA	72.3	258.0	202976*	0.76	1.91	0.91	2.11
22.9	12.4	18,22,27,31,36,40, 44,53,62	2	62.3	224.8	505025	1.19	2.54	1.60	3.86
22.9	12.4	53,58	2	57.8	208.8	449915	1.17	2.51	1.55	2.03
22.9	12.4	49,58,62,67	3	66.7	240.1	944166	1.14	2.64	1.55	4.11
22.9	12.4	53,62	3	62.3	225.0	994876	1.17	2.64	1.80	1.98
22.9	12.4	67	NA	66.7	240.6	425378*	0.71	1.83	1.55	2.90
22.9	12.4	44,53	3	53.4	192.6	1660282	1.37	2.82	1.85	4.42
22.9	13.0	58	NA	57.8	203.5	1037016*	0.86	1.91	3.94	6.99

*first step initiation, **spot weld initiation

Table 3 Test results for shot peened aluminum 7075-T7351 corrosion pit specimens.

<i>w</i> mm	<i>t</i> mm	<i>load steps</i> kN	<i>Cycles</i> /step E6	<i>final</i> <i>P</i> _{max} kN	<i>final</i> <i>S</i> _{max} MPa	<i>final step</i> cycles	<i>a</i> _i mm	<i>2c</i> _i mm	<i>a</i> _f mm	<i>2c</i> _f mm
Nominal Shot Peen Aluminum 7075-T7351 R=0.1										
23.4	12.4	40	NA	40.0	141.7	89045*	0.84	1.85	1.98	4.19
23.4	12.4	44	NA	44.5	157.2	77107*	0.74	1.98	1.85	4.27
23.4	12.4	27	NA	26.7	94.4	251863*	0.94	2.11	1.60	3.15
22.9	12.4	18,22,27	3	26.7	95.5	6377641	0.86	1.78	1.50	3.02
22.9	12.4	18	NA	17.8	64.1	276391*	1.27	2.90	1.55	3.66
22.9	12.4	13	NA	13.3	48.1	49893*	1.19	2.92	2.01	3.89
22.9	12.4	22,27,31,36,40	3	40.0	143.9	262497	0.74	1.68	1.68	3.02
22.9	12.4	18,22,27,31,36	3	35.6	127.7	392494	0.71	1.70	1.07	2.46
22.9	12.4	27,31	3	31.1	108.6	498885	0.66	1.40	0.84	1.78
22.9	12.4	27,31,36	3	35.6	128.7	523297	0.69	1.88	2.41	5.79
22.9	12.4	18,22,27	3	26.7	96.2	573200	1.24	3.23	3.00	4.88
Nominal Shot Peen Aluminum 7075-T7351 R=0.7										
23.4	12.4	22,27,31,36, 40,44,49,53	2	53.4	192.6	796450	0.86	1.55	1.17	2.24
22.9	12.4	36,40,44, 49,53	2	53.4	194.2	616643	0.89	2.11	1.60	3.38
22.9	12.4	36,44,53	3	53.4	193.9	3000000	1.24	3.07	2.64	4.52
22.9	12.4	36,44,53	3	53.4	191.7	418230	1.35	3.07	2.11	4.34
*first Step initiation										

Table 4 Test results for heavy shot peened aluminum 7075-T7351 corrosion pit specimens.

<i>w</i> mm	<i>t</i> mm	<i>load steps</i> kN	<i>Cycles/</i> <i>step</i> <i>E6</i>	<i>final</i> <i>P_{max}</i> kN	<i>final</i> <i>S_{max}</i> MPa	<i>final</i> <i>step</i> <i>cycles</i>	<i>a_i</i> mm	<i>2c_i</i> mm	<i>a_f</i> mm	<i>2c_f</i> mm
Heavy Shot Peen Aluminum 7075-T7351 R=0.1										
22.9	12.7	22	3	22.2	77.8	582909*	1.73	3.07	3.86	3.99
22.9	12.7	13,18,22,27	3	26.7	93.5	373952	1.70	3.51	3.99	4.32
23.1	12.7	18,22	3	22.2	77.4	1044904	1.73	3.28	4.88	5.51
23.1	12.7	18,22	3	22.2	77.4	439232	1.80	3.58	4.67	5.11
23.1	12.7	18,22	3	22.2	77.4	611853	1.75	3.73	4.19	4.06
23.1	12.7	9,13,18,22, 27,31	3	31.1	108.4	244487	1.75	3.76	3.45	3.30
23.1	12.7	18,22,27, 31,36	3	35.6	123.9	305803	1.30	3.07	3.23	3.76
22.9	13.0	18	3	31.1	108.6	244192*	1.32	3.12	2.95	3.86
23.1	13.0	24,31,36	3	22.2	77.2	261658	1.40	2.90	4.34	6.02
22.9	12.7	27,31	3	31.1	109.1	784017	1.35	3.15	3.61	3.30
22.9	12.7	27,31	3	31.1	109.3	340689	1.42	2.82	3.96	4.01
22.9	12.7	27,31,36	3	35.6	125.0	325466	1.30	2.54	3.84	3.91
Heavy Shot Peen Aluminum 7075-T7351 R=0.7										
22.9	12.7	31,36,40,44, 49,58	3	57.8	202.2	612040	2.03	3.33	2.29	0.00
22.9	12.7	40,44,	3	44.5	155.6	2280781	1.78	3.30	3.23	4.29
22.9	12.7	40,44,49,	3	44.5	155.6	848822	1.83	3.23	3.43	4.90
22.9	12.7	40,44,	3	44.5	155.6	1133015	1.80	3.45	2.95	4.75
22.9	13.0	40,44,49	3	48.9	172.2	1119531	1.70	3.30	2.87	4.60
22.9	12.7	38,42	3	42.3	148.1	1530802	1.40	2.79	3.51	3.43
22.9	12.7	44,49	3	48.9	171.5	679779	NA	2.90	2.64	2.57
22.9	12.7	44,49	2	48.9	171.8	1369454	1.42	2.84	2.41	2.24
22.9	12.7	44,49,53,58	5	66.7	234.3	440039	1.35	2.67	3.48	4.04
25.7	12.9	49	3	48.9	152.8	2219412	1.40	3.25	3.00	3.53
*first step initiation										

Table 5 Test results for bare 4340 steel corrosion pit specimens.

<i>w</i> mm	<i>t</i> mm	<i>load steps</i> kN	<i>Cycles</i> /step E6	<i>final</i> <i>P</i> _{max} kN	<i>final</i> <i>S</i> _{max} MPa	<i>final step</i> cycles	<i>a</i> _i mm	<i>2c</i> _i mm	<i>a</i> _f mm	<i>2c</i> _f mm
4340 Steel Bare R=0.1										
22.9	11.4	111	NA	111.2	437.5	491560*	0.99	2.16	1.27	3.48
22.9	11.4	100	NA	100.1	396.9	237177*	0.94	1.93	1.60	3.18
22.9	11.4	80,89,98,107	2	106.8	422.3	405660	0.91	1.19	1.07	1.52
22.9	11.4	89,102,116,129	2	129.0	510.3	88400	1.55	2.54	1.80	3.89
23.1	11.4	102,116,129	2	129.0	517.2	76264	1.35	2.44	1.91	4.22
23.1	11.4	89,102,116	2	115.6	463.7	155389	1.57	2.57	1.60	3.91
22.9	11.2	89,98,111	3	111.2	450.5	935503**	0.64	1.83	NA	NA
22.9	11.2	89,98,111	3	111.2	446.9	1464655	0.58	1.78	1.37	3.25
23.1	11.2	89,98,111,124	2	124.5	498.2	285149	0.58	2.06	1.17	3.25
4340 Steel Bare R=0.7										
22.9	11.4	124,142,156,169	3	169.0	674.0	285076	0.97	2.39	1.42	3.58
22.9	11.2	142,156,169,187, 205,222,242	2	242.4	972.0	1747582	1.47	2.49	2.51	3.25
22.9	11.2	147,164,182, 200,218,240	2	240.2	965.3	207141**	0.69	1.32	NA	NA
22.9	11.2	142,156,169, 187,200,214	3	213.5	860.0	1014403 ⁺	0.66	1.83	NA	NA
22.9	11.2	156,169,182,200	2	200.2	803.5	427981 ⁺⁺	0.99	1.93	NA	NA
22.9	11.4	178,196,214,236,	2	235.7	928.4	219982	0.58	1.17	1.42	1.73
22.9	11.4	178,196,214,231	3	231.3	926.4	198006	0.69	1.93	1.09	2.24
22.9	11.4	160,178,205,222	2	222.4	877.0	429015	0.69	2.13	1.65	3.40

*first step initiation, **spot weld initiation, ⁺specimen propagated to fracture, ⁺⁺hydraulic overload

Table 6 Test results for shot peened 4340 steel corrosion pit specimens.

<i>w</i> mm	<i>t</i> mm	<i>load steps</i> kN	<i>Cycles</i> /step E6	<i>final</i> <i>P</i> _{max} kN	<i>final</i> <i>S</i> _{max} MPa	<i>final step</i> cycles	<i>a</i> _i mm	<i>2c</i> _i mm	<i>a</i> _f mm	<i>2c</i> _f mm
4340 Steel Shot Peen R=0.1										
22.9	11.4	107,120,133,147	2	146.8	581.4	109104	1.27	2.54	2.01	4.09
22.9	11.4	133	NA	133.4	528.5	115860*	1.27	2.54	1.93	3.84
22.9	11.4	133	NA	133.4	528.5	98776*	1.22	2.46	1.93	1.14
22.9	11.4	98,107,116, 124,138	2	137.9	546.1	435489	1.17	2.39	2.01	3.66
22.9	11.4	147	NA	146.8	576.8	104055*	1.24	2.46	1.70	3.51
22.9	11.4	178	NA	177.9	699.3	40957*	0.71	1.93	1.70	2.49
22.9	11.4	133,147,160	2	160.1	634.2	69648	0.58	1.37	1.70	2.26
22.9	11.4	133,147	3	146.8	581.4	161259	0.56	1.17	1.32	1.80
22.9	11.4	133	NA	133.4	528.5	548291**	0.58	1.24	NA	NA
4340 Steel Shot Peen R=0.7										
22.9	11.4	169,187,205,222, 245,267,289	2	289.1	1,145.1	90243	1.19	2.46	1.65	2.77
22.9	11.4	205,222,245,	2	244.6	968.9	348067	1.17	2.06	1.85	2.06
22.9	11.4	205,222,245,267	2	266.9	1,057.0	43806	1.14	2.36	2.16	2.95
22.9	11.5	178,196	2	195.7	767.3	1965058	0.79	1.80	NA	NA
22.9	11.4	222,245,267	2	266.9	1,057.0	318476	0.61	1.22	1.07	1.40

*first step initiation, **spot weld initiation

Table 7 Test results for laser peened 4340 steel corrosion pit specimens.

<i>w</i> mm	<i>t</i> mm	<i>load steps</i> kN	<i>Cycles</i> /step E6	<i>final</i> <i>P</i> _{max} kN	<i>final</i> <i>S</i> _{max} MPa	<i>final step</i> <i>cycles</i>	<i>a</i> _i mm	<i>2c</i> _i mm	<i>a</i> _f mm	<i>2c</i> _f mm
4340 Steel Laser Peen, R=0.1										
22.9	11.4	169,187,205,222, 245,267,289	2	289.1	1,145	90243	1.19	2.46	1.65	2.77
22.9	11.4	205,222,245,	2	244.6	968.9	348067*	1.17	2.06	1.85	2.06
22.9	11.4	205,222,245,267	2	266.9	1,057	43806^	1.14	2.36	2.16	2.95
22.9	11.5	178,196	2	195.7	767.3	1965058*^	0.79	1.80	NA	NA
22.9	11.4	222,245,267	2	266.9	1,057	318476*	0.61	1.22	1.07	1.40
4340 Steel Laser Peen, R=0.7										
22.9	11.4	89,97,111,124, 138,151,169	2	169.0	669.4	172034	1.22	2.59	2.06	3.81
22.9	11.4	178	NA	177.9	704.6	731953	1.17	2.57	3.53	4.60
22.9	11.4	151,169	2	169.0	669.4	313443	1.24	2.64	NA	NA
22.9	10.9	151	NA	151.2	619.0	486065	0.79	2.44	NA	NA
22.9	10.9	156,169	2	169.0	697.5	35096	0.71	1.96	NA	NA

*first step initiation, ^cracking away from pit

Table 8 Test results for bare D6AC steel corrosion pit specimens.

<i>w</i> mm	<i>t</i> mm	<i>load steps</i> kN	<i>Cycles</i> /step E6	<i>final</i> <i>P</i> _{max} kN	<i>final</i> <i>S</i> _{max} MPa	<i>final step</i> <i>cycles</i>	<i>a</i> _i mm	<i>2c</i> _i mm	<i>a</i> _f mm	<i>2c</i> _f mm
D6AC Steel Bare, R=0.1										
22.9	10.2	80,84,89,98, 107	2.5	106.8	471.9	255223	0.61	1.73	1.14	2.64
22.9	7.9	80,76,80,84,93		80.1	456.7	175540*	0.58	1.65	1.27	2.59
22.9	8.9	49,58,67,76,84	2.5	93.4	467.9	337176	0.64	1.98	1.24	2.54
22.9	9.9	71,76,80,87,	3	84.5	379.7	345037	1.17	2.54	3.58	6.17
22.9	10.2	72,76,80,87	3	86.7	383.0	152714	1.14	2.51	1.40	3.38
22.9	9.9	72,76,80,84	3	84.5	383.2	521214	1.04	2.92	2.06	4.80
22.9	9.9	71,76,80,85, 102	3	102.3	465.1	114099	1.17	2.64	1.60	3.38
22.9	9.7	80,89	3	89.0	416.2	220434	1.17	2.79	1.73	4.06
22.9	10.2	89		89.0	395.2	441687*	1.17	2.64	2.16	3.81
D6AC Steel Bare, R=0.1										
22.9	9.7	89,98,107,120, 133,147,160, 178,196	3	195.7	911.1	293769	0.58	1.47	1.22	2.26
22.9	10.2	147,160,173,191, 209	3	209.1	915.0	501709	0.56	1.70	1.50	2.95
22.9	10.2	160,173,191,209	3	209.1	935.9	334648	0.66	1.68	1.37	2.39
22.9	9.9	178,196	3	195.7	884.1	442707	0.64	1.57	1.30	2.54
22.9	9.9	177,196,214	3	213.5	954.7	644712	0.61	1.63	0.99	2.39
22.9	9.9	178,196,	3	195.7	890.9	515419	0.64	1.57	1.07	2.49
22.9	10.2	133,147,160,178, 196	3	195.7	865.2	1197348	1.32	2.90	NA	NA
22.9	10.2	160,178,196,214	3	213.5	943.9	172600	1.22	2.82	1.65	3.28
22.9	10.4	178,196,214	3	213.5	929.9	277400	1.19	2.64	2.03	2.21

*first step initiation

Table 9 Test results for shot peened D6AC steel corrosion pit specimens.

<i>w</i> <i>mm</i>	<i>t</i> <i>mm</i>	<i>load steps</i> <i>kN</i>	<i>Cycles</i> <i>/step</i> <i>E6</i>	<i>final</i> <i>P_{max}</i> <i>kN</i>	<i>final</i> <i>S_{max}</i> <i>MPa</i>	<i>final step</i> <i>cycles</i>	<i>a_i</i> <i>mm</i>	<i>2c_i</i> <i>mm</i>	<i>a_f</i> <i>mm</i>	<i>2c_f</i> <i>mm</i>
D6AC Steel Shot Peened, R=0.1										
22.9	11.4	89,102,116,129	2	129.0	509.2	117366	1.17	2.39	1.93	3.25
22.9	10.7	167,120,133	3	133.4	561.2	117146	1.19	2.44	1.96	3.76
22.9	11.2	107	NA	0.1	431.5	200547*	1.14	2.41	1.63	3.73
22.9	11.2	93,107	3	0.1	428.6	199610	1.27	2.62	1.50	1.65
22.9	10.4	80,89,98,107	3	106.8	462.1	135610	1.22	2.87	1.83	5.49
		89,98,107,116, 124,133,142,								
22.9	11.4	151,160,173	3	173.5	682.4	79233	1.17	2.41	1.52	3.45
22.9	11.4	98,111,	2	111.2	439.9	210390	0.71	2.31	1.75	2.90
23.1	11.4	89	NA	89.0	350.3	231400*	0.61	1.40	1.32	2.79
D6AC Steel Shot Peened, R=0.7										
22.9	11.2	160,173,187, 200,214,222	3	222.0	898.1	4000000>	1.24	2.39	NA	NA
22.9	11.2	160,173,187, 205,220,245	2	244.6	992.2	200793	1.14	2.26	1.60	3.51
22.9	11.4	160,169	2	169.0	663.4	913680	1.19	2.49	1.91	3.58
22.9	11.4	160,173,187, 205,222,240	2	240.2	944.9	114308	1.22	2.39	1.63	3.30
22.9	11.4	178,196,214,236,	2	235.7	932.6	2000000	0.61	2.26	0.97	2.95
22.9	11.4	205,222	2	222.4	866.1	258696	0.64	1.32	1.37	1.63
22.9	11.4	160		160.1	632.1	661961	0.58	1.22	1.37	2.69
22.9	11.2	178,196,214, 231,240,267	3	266.9	1083.6	1699	0.58	1.22	0.94	2.29
*first step initiation, >aborted test										

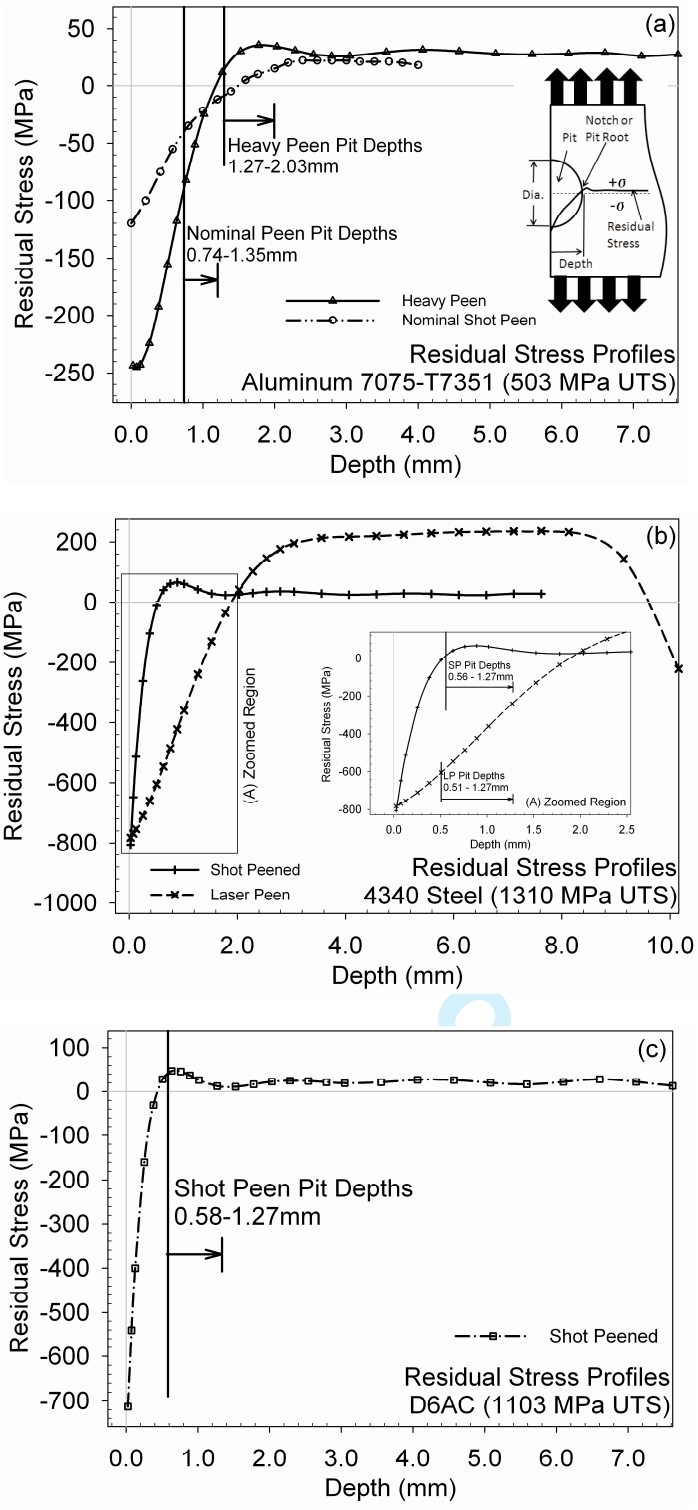


Fig. 1 Residual stress measurements for the aluminum 7075-T7351 (a), 4340 steel (b), and D6AC steel (c).

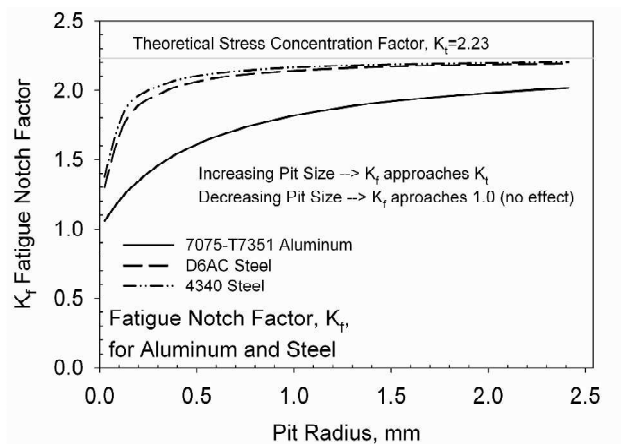


Fig. 2 Fatigue notch factors as a function of pit radius for aluminum and steel alloys.

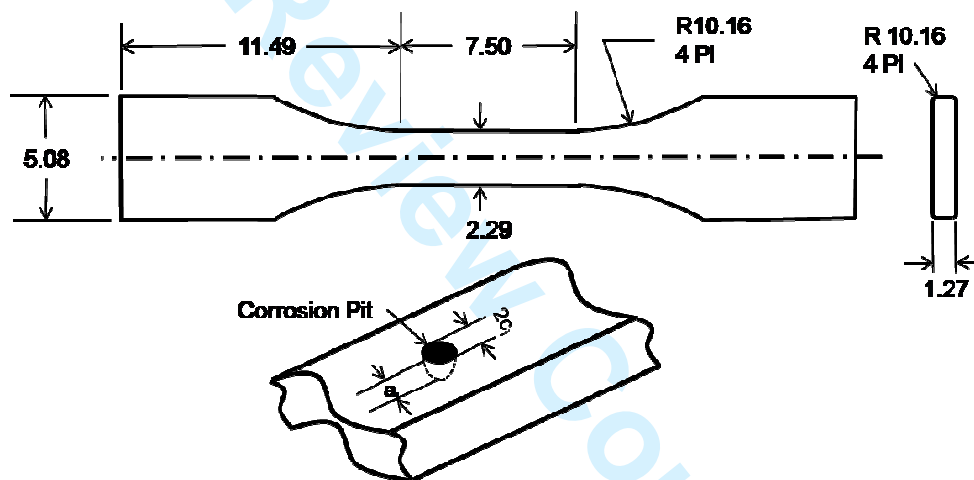


Fig. 3 Specimen schematic and reference corrosion pit location.

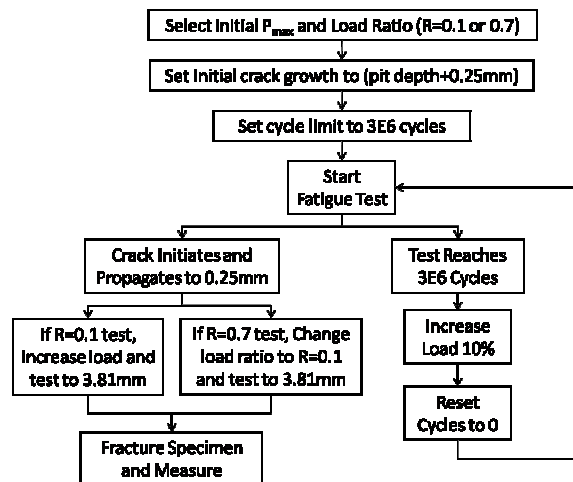


Fig. 4 Fatigue crack initiation and propagation step test process.

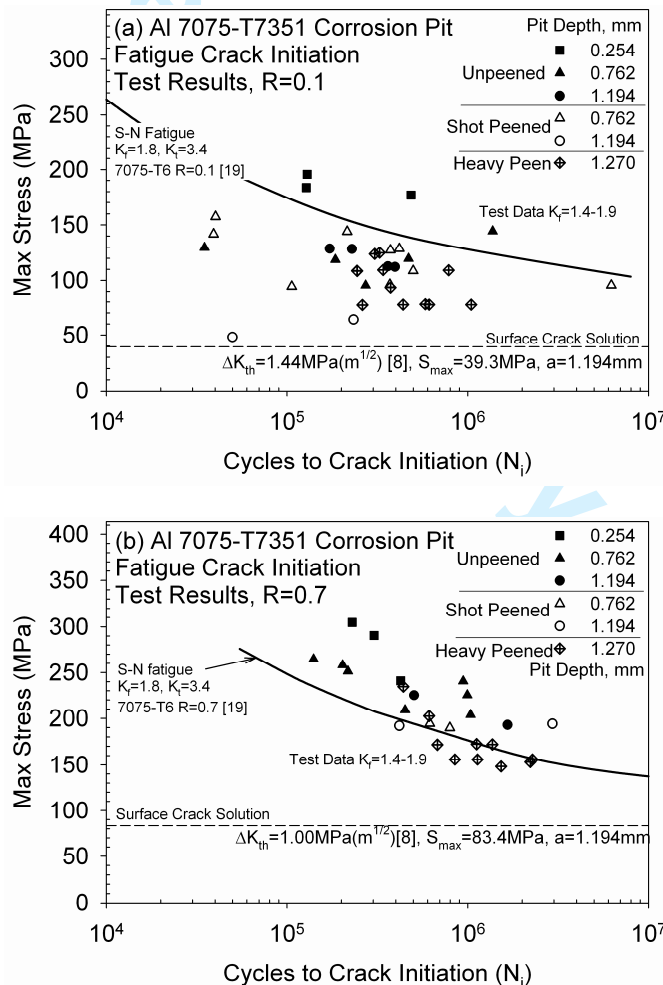


Fig. 5 Al 7075-T7351 test results for bare and nominal shot peened specimens, R=0.1.

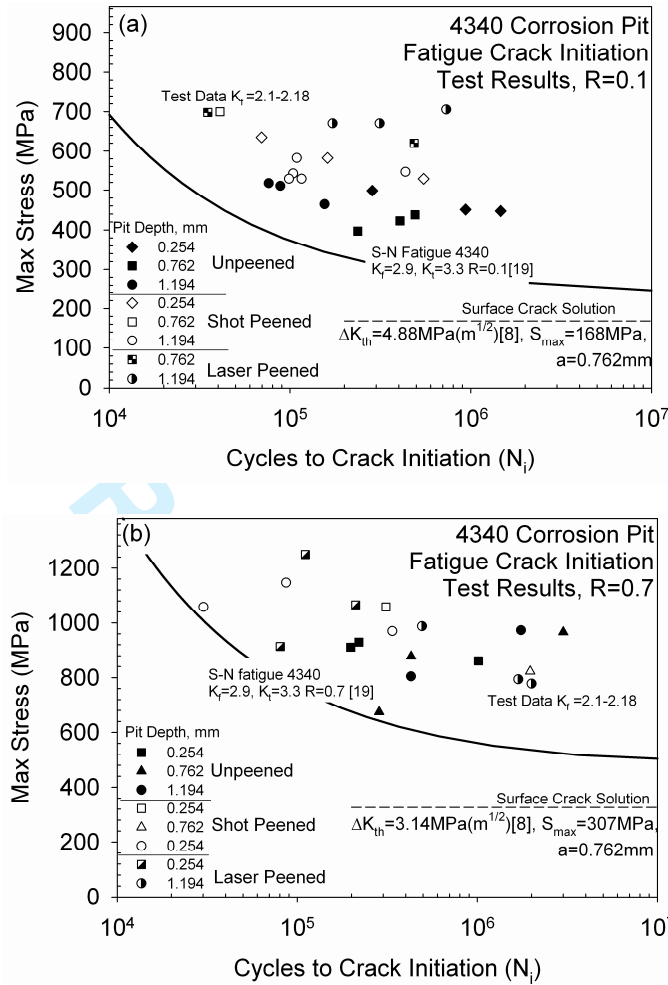


Fig. 6 4340 steel fatigue test results for all surface conditions, R=0.1.

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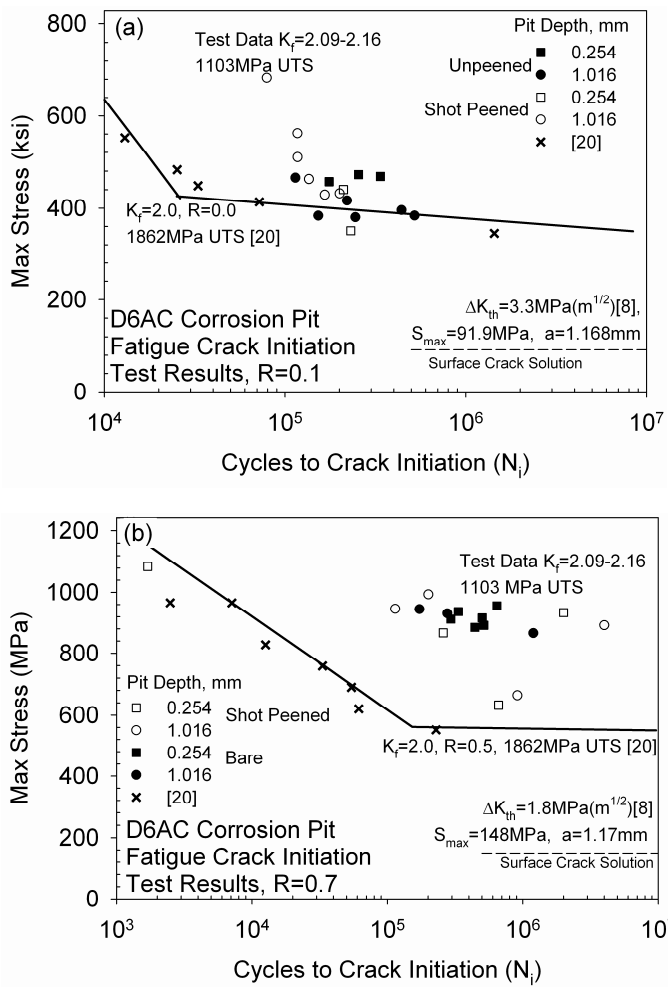


Fig. 7 D6AC steel bare and shot peened test results for R=0.1.

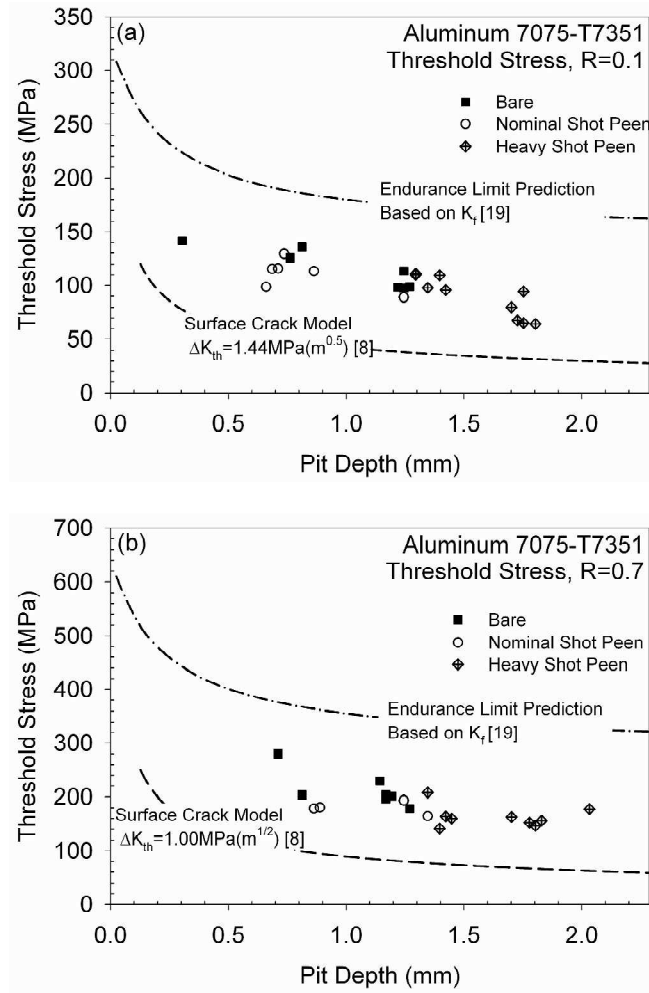


Fig. 8 Threshold stress as a function of pit depth for aluminum 7075-T7351.

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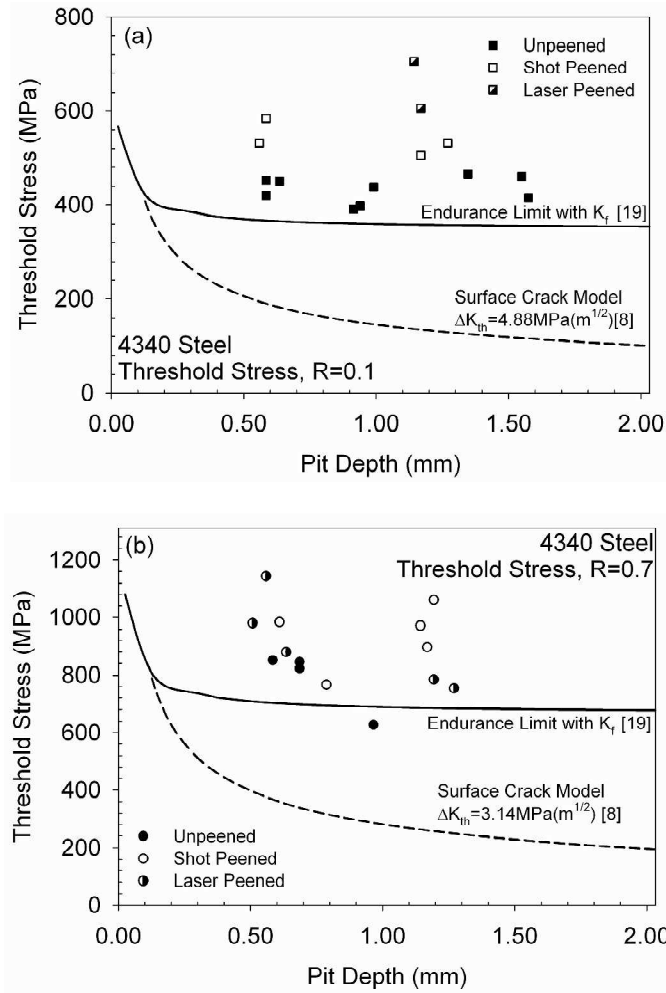


Fig. 9 Threshold stress for the 4340 steel as a function of pit depth.

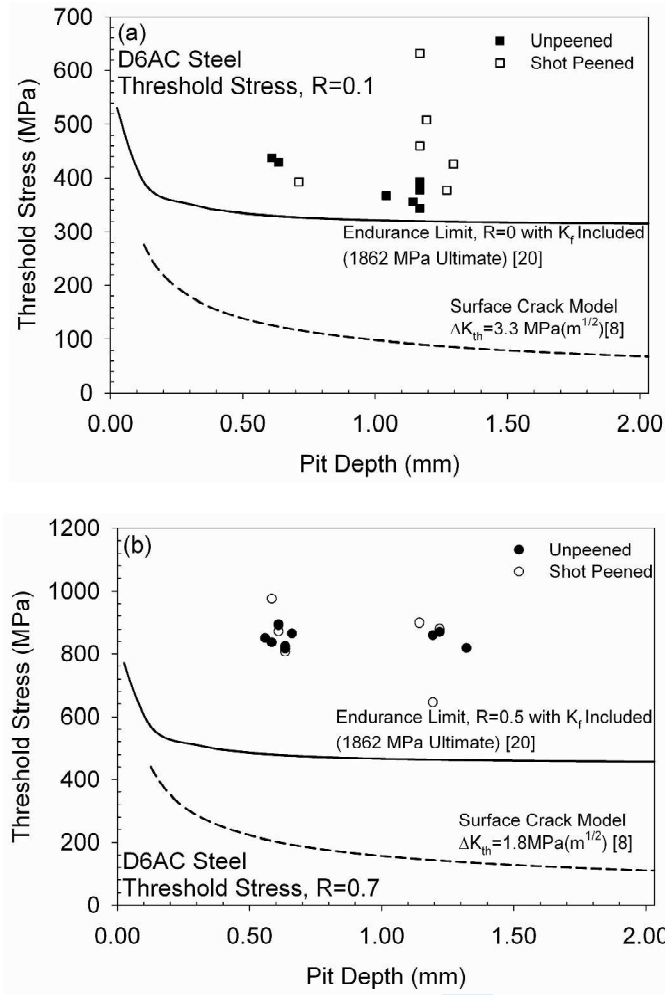


Fig. 10 Threshold stress for the D6AC steel as a function of pit depth.