

Heating Augmentation for Short Hypersonic Protuberances

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Computational aeroheating analyses of the Space Shuttle Orbiter plug repair models are validated against data collected in the Calspan University of Buffalo Research Center (CUBRC) 48 inch shock tunnel. The comparison shows that the average difference between computed heat transfer results and the data is about 9.5%. Using CFD and Wind Tunnel (WT) data, an empirical correlation for estimating heating augmentation on short hypersonic protuberances ($k/\delta < 0.3$) is proposed. This proposed correlation is compared with several computed flight simulation cases and good agreement is achieved. Accordingly, this correlation is proposed for further investigation on other short hypersonic protuberances for estimating heating augmentation.

Nomenclature

<i>Symbols</i>		θ	Momentum thickness, [<i>inches</i>]
C_p	Specific heat, [<i>BTU/slugsR</i>]	<i>Subscripts</i>	
d	Plug repair diameter, [<i>inches</i>]	∞	Free-stream
k	Protuberance height, [<i>inches</i>]	i, j, k	Grid points indices
M	Mach number	p	Plug repair
p	Pressure, [<i>lb/ft²</i>]	rf	Recover factor
q	Heat transfer rate, [<i>BTU/ft² · sec</i>]	t	Total
r	Constant	θ	Momentum thickness
Re	Unit Reynolds number, [<i>1/ft</i>]	w	Wall
St	Stanton number	x, y, z	Local x-, y-, z-coordinate
T	Temperature, [<i>R</i>]	$\langle . \rangle$	Average value
V	Total velocity, [<i>ft/sec</i>]	<i>Acronyms</i>	
x, y, z	Coordinate system, [<i>inches</i>]	BF	Bump factor
α	Angle-of-attack, [<i>degrees</i>]	BL	Boundary layer
δ	BL thickness, [<i>inches</i>]	WLE	Wing leading edge
δ^*	Displacement thickness, [<i>inches</i>]	WT	Wind tunnel
ϵ	Emissivity		
μ	Viscosity, [<i>lb · sec/ft²</i>]		
ρ	Density, [<i>slugs/ft³</i>]		

Introduction

Following the Shuttle Columbia accident in 2003, NASA investigated methods for performing on-orbit repairs to the Thermal Protection System (TPS) and their impacts on the aerothermodynamic environment of the Space Shuttle Orbiter experienced during re-entry.¹ Various methods were developed to repair damaged areas of the Space Shuttle Orbiter, including the use of tile repairs and plug repairs that are used on-orbit to patch small breaches on the Orbiter tile and on the Wing Leading Edge (WLE) Reinforced Carbon-Carbon (RCC) panels, respectively. Several studies have been reported for the aeroheating analysis of the tile and

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plug repairs (see Ref.²⁻⁷) However, no correlation has been reported to estimate the augmented maximum heating resulting from disturbing the local flow field.

The aim of this investigation is to obtain a heating augmentation correlation for a hypersonic repair surface using flow field information around the smooth Outer Mold Line (OML) of the vehicle. The focus of this study is on the Space Shuttle Orbiter. Aerothermodynamic computations are studied for wind tunnel swept cylinder models with rounded protuberance of different heights is placed on the cylinder. CFD simulations are obtained for the WT cases and the results are compared with the WT data. Based on CFD and WT data, a simple heating augmentation correlation is proposed for comparison with several flight CFD data. The correlation does require that Shuttle Smooth OML results are available.

CFD Validation

In this section, computational analysis are validated against available wind tunnel data. The validation is necessary in order to extrapolate the CFD data to flight conditions for which a heating correlation augmentation is needed.

Computational Procedures

Computational aerothermodynamic analyses are conducted using Langley Aerothermodynamic Upwind Relaxation Algorithm (LAURA),^{8,9} which is a chemically reacting viscous flow solver. Computations are performed for three-dimensional swept cylinder models, which are shown schematically in Figures 1 and 2.

Several different plug heights are considered. The plug dimensions and wind tunnel flow conditions corresponding to the experimental run numbers are given in Table 1. Perfect gas air is used during the experiments. Experiments are conducted by Calspan University of Buffalo Research Center (CUBRC)¹⁰ in a 48 *inch* shock tunnel. Computations are simulated at the wind tunnel conditions with a perfect gas air model.

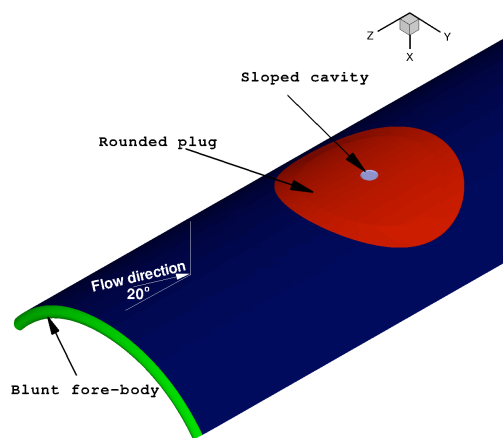


Figure 1. Schematic of the wind tunnel model.

Table 1. Plug dimensions and wind tunnel flow conditions corresponding to the experimental run numbers.

Run Number	d_p inches	k_p inches	M_∞	V_∞ ft/sec	$\rho_\infty \times 10^6$ slugs/ft ³	T_∞ R	$Re \times 10^{-6}$ 1/ft	k/δ
15	7	0.025	11.60	6290	2.86	122	0.19	0.11
13	7	0.045	11.57	6360	2.80	126	0.18	0.20
28	7	0.055	11.68	6270	2.77	120	0.19	0.24
11	7	0.065	11.68	6190	2.77	117	0.19	0.28

Because each WT model was individually manufactured, a specific grid was needed for each of the models to account for their precise features. A grid topology was designed to generate smooth grids that can capture geometrical features of the model and physical behavior of hypersonic flow around the body. A schematic of the developed viscous grid is shown in Figure 3. More details of the grid generating techniques are reported in Ref.¹²

Computational Results

Wind tunnel models, each consisting of a 7 *inch* rounded plug on a swept cylinder with a blunted nose, shown in Figures 1 and 2, are analyzed at the wind tunnel conditions given in Table 1. As stated before, perfect gas air is used for both wind tunnel experiments and computations. Simulations are conducted with

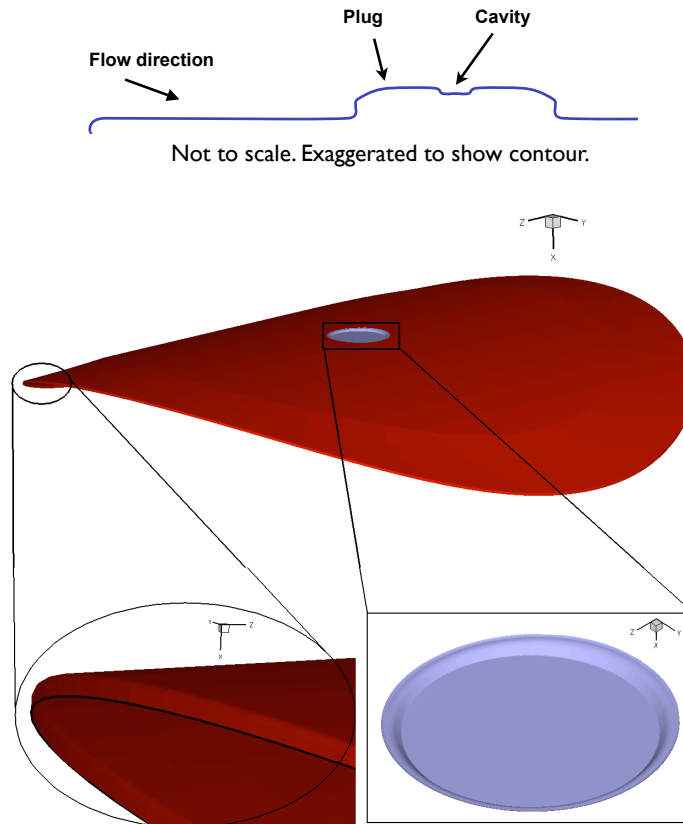


Figure 2. Schematic of the plug shape with its cavity.

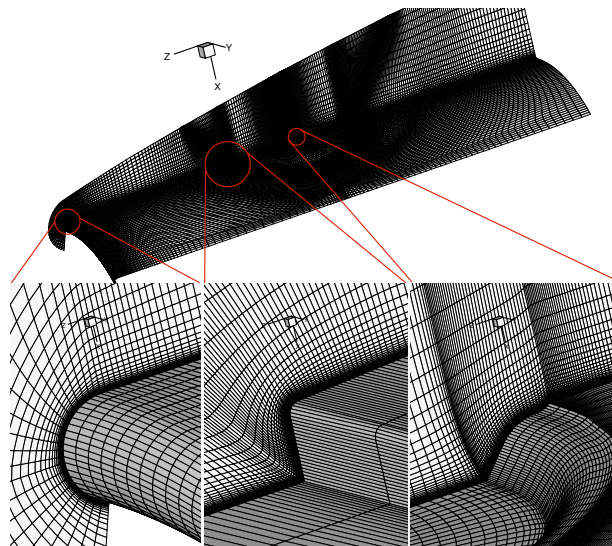


Figure 3. Viscous grid developed for a wind tunnel swept cylinder model.

the WT conditions for which the surface temperatures are relatively constant (cold wall). WT conditions are considered laminar and therefore only laminar flow is computed.

Contour plots of the computed surface heat fluxes are shown in Figure 4. This figure shows that the plug surface heat flux increases with the plug height. The local surface heat flux is employed and local surface Stanton number is calculated.

The main goal of this study is to obtain a correlation that may be used for flight conditions in which the surface is at radiative equilibrium with surface emissivity of $\epsilon = 0.89$. Consequently, the Stanton number is obtained for radiative equilibrium surface conditions. The Stanton number is calculated as

$$St = q_w / (\rho_\infty V_\infty C_p (r_{rf} T_t - T_w)) \quad (1)$$

where q_w is surface heat flux, ρ_∞ is free-stream density, V_∞ is free-stream velocity, C_p is specific heat at constant pressure, r_{rf} is a constant recovery factor, T_t is total temperature, and T_w is surface temperature. The values of free-stream density and velocity are given in Table 1, while values of specific heat, total temperature and wall temperature are given in Table 2. A constant recovery factor of 0.92 is used for all the calculations.¹³ The Surface Stanton number variations are plotted in Figure 5.

Specific heat values are calculated based on wind tunnel conditions. For the radiative equilibrium wall condition, local surface temperature is used. Figure 5 shows that the local Stanton number computed with the radiative equilibrium wall temperature boundary condition does not entirely match the one computed with the cold wall boundary condition. The difference is largest at the plug leading edge, and becomes comparable far upstream and downstream of the plug.

For all WT cases, the flow spreads to both sides of the plug before passing the plug leading edge and causing a small recirculation region upstream of the plug. Streamlines on the plug top surface are similar with a recirculation zone just downstream of the cavity trailing edge. These flow features are schematically shown in Figure 6 for the 0.065 inch plug.

Computational Validation

The CFD analyses are validated by comparing the numerical data with the WT data. The surface heat flux probe locations are shown in Figure 7. Table 3 summarizes differences computational and experimental¹⁰ surface heat flux data for all gauge locations. The average of absolute percentage errors are also given in the last column. This table shows that the averaged data from the CFD simulation are within 12% of the experimental data except for the probe number 7 where the average difference is about 30%. Probe 7 is located at the base of the plug where it meets the swept cylinder, and a large error is possibly due to modeling limitation in such area. The average difference between computed and reported WT heat transfer data is about 8% and 9.5% with and without considering probe 7, respectively.

The accuracy of the numerical results are emphasized in Figure 8 by showing heat flux variations with distance along a line that connects two probe points of 1 and 16 (see Figure 7a.) Wind tunnel data are also

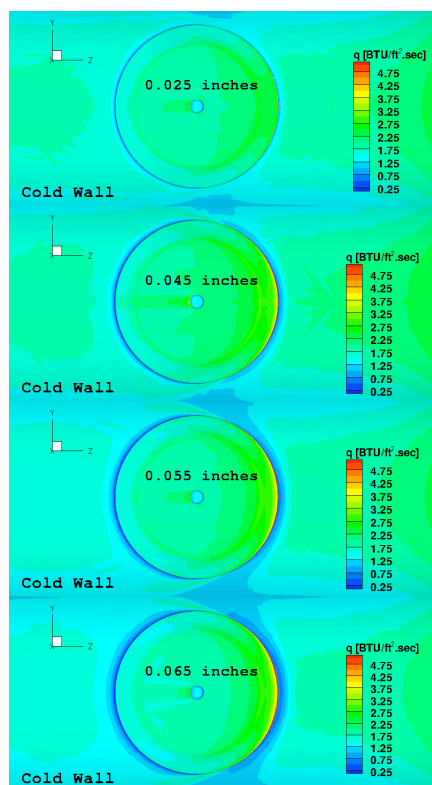


Figure 4. Computed surface heat flux variations on the plug area of the wind tunnel geometries. Flow direction is from right to left.

Table 2. Parameters used in the Stanton number calculations.

k_p inches	C_p BTU/slugs · R	T_w R	T_t R
0.025	7.7247	526	3110
0.045	7.7081	526	3170
0.055	7.7286	521	3090
0.065	7.7519	525	3020

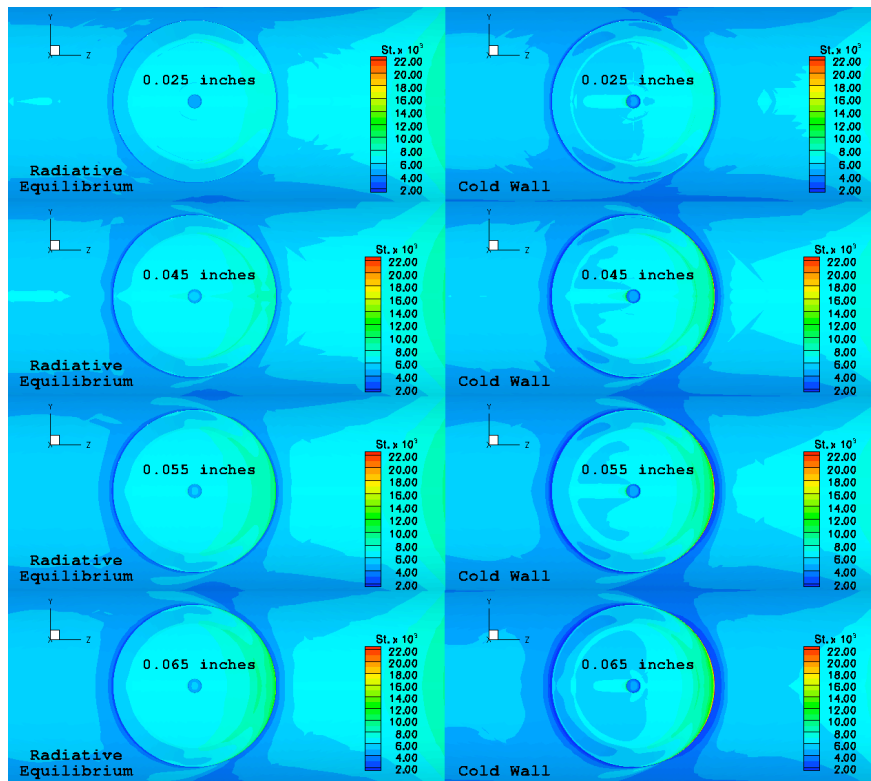


Figure 5. Stanton number variations on the plug area of the wind tunnel geometries. Flow direction is from right to left.

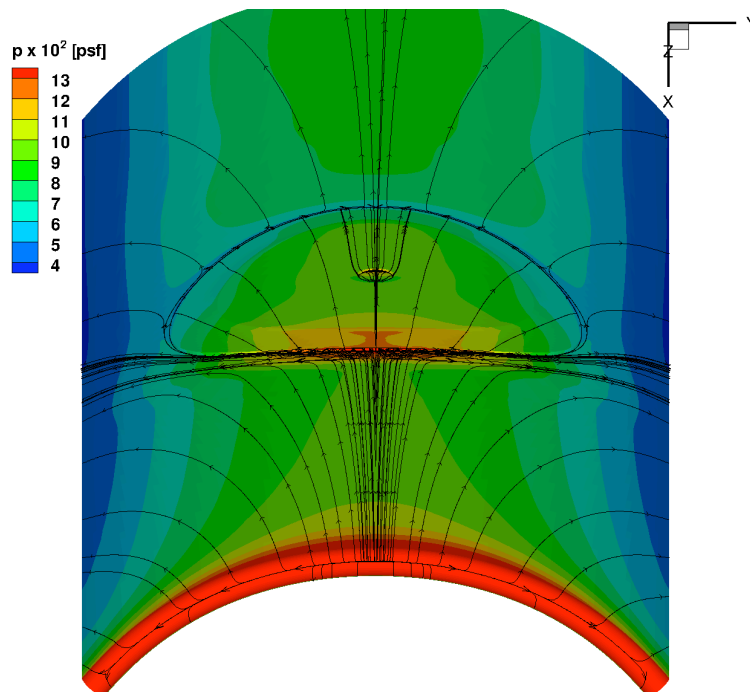


Figure 6. Streamtraces of flow around the the 0.065 inch wind tunnel geometry.

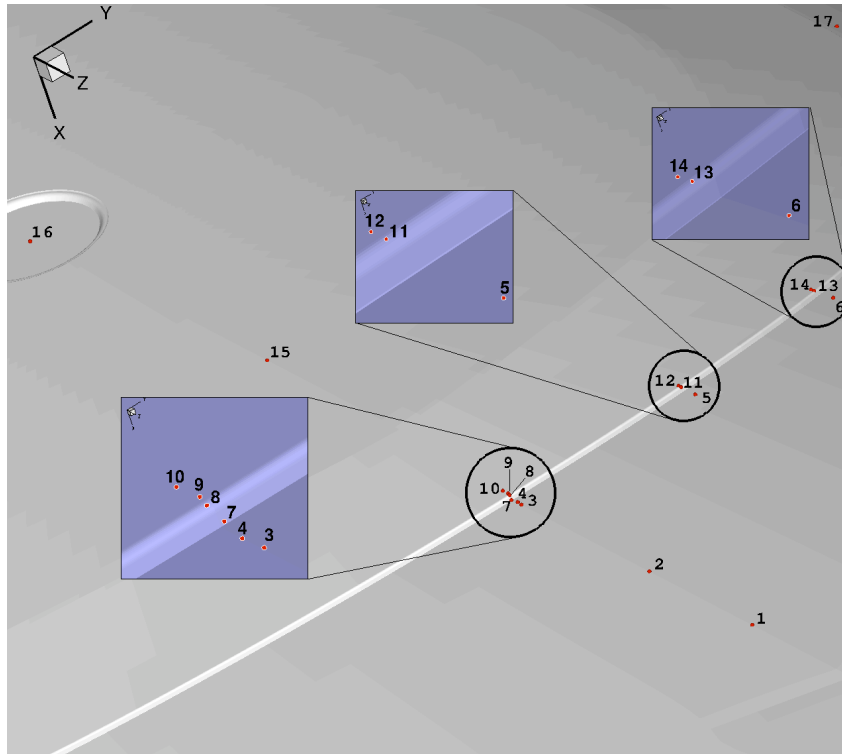


Figure 7. Surface data points.

Table 3. Differences between computed and measured surface heat flux.

DATA POINT	<i>Error</i>				$< Error >$
	%				
	<i>0.025 inches</i>	<i>0.045 inches</i>	<i>0.055 inches</i>	<i>0.065 inches</i>	
1	-2.7	+3.0	-1.5	-5.2	3.1
2	-6.8	+13.2	+8.53	+4.4	8.2
3	+17.5	-0.6	+12.4	-12.6	10.8
4	+11.5	+5.9	-10.2	-19.3	11.7
5	+16.9	+1.9	+4.3	-12.8	9.0
6	+13.0	+4.7	+13.0	-2.5	8.3
7	-58.6	+21.0	+34.0	+9.2	30.7
8	-11.8	+24.3	-3.7	-1.6	10.3
9	-11.6	+19.2	+5.6	+7.0	10.8
10	-5.4	+18.9	+11.9	+10.9	11.8
11	+3.3	+19.6	+1.9	-6.5	7.8
12	+0.7	+8.4	+12.0	+3.3	6.1
13	-10.2	+2.4	+13.1	+0.2	6.5
14	-13.1	-	-3.4	-8.1	8.2
15	+4.8	13.2	+7.3	+0.8	6.5
16	-11.3	+4.9	+0.4	-6.8	5.9
17	-3.6	+6.4	+6.7	+3.8	5.1

shown in this figure with plug geometry variations. A very good agreement with WT data is achieved. This figure shows that the peak heating increases with the plug height.

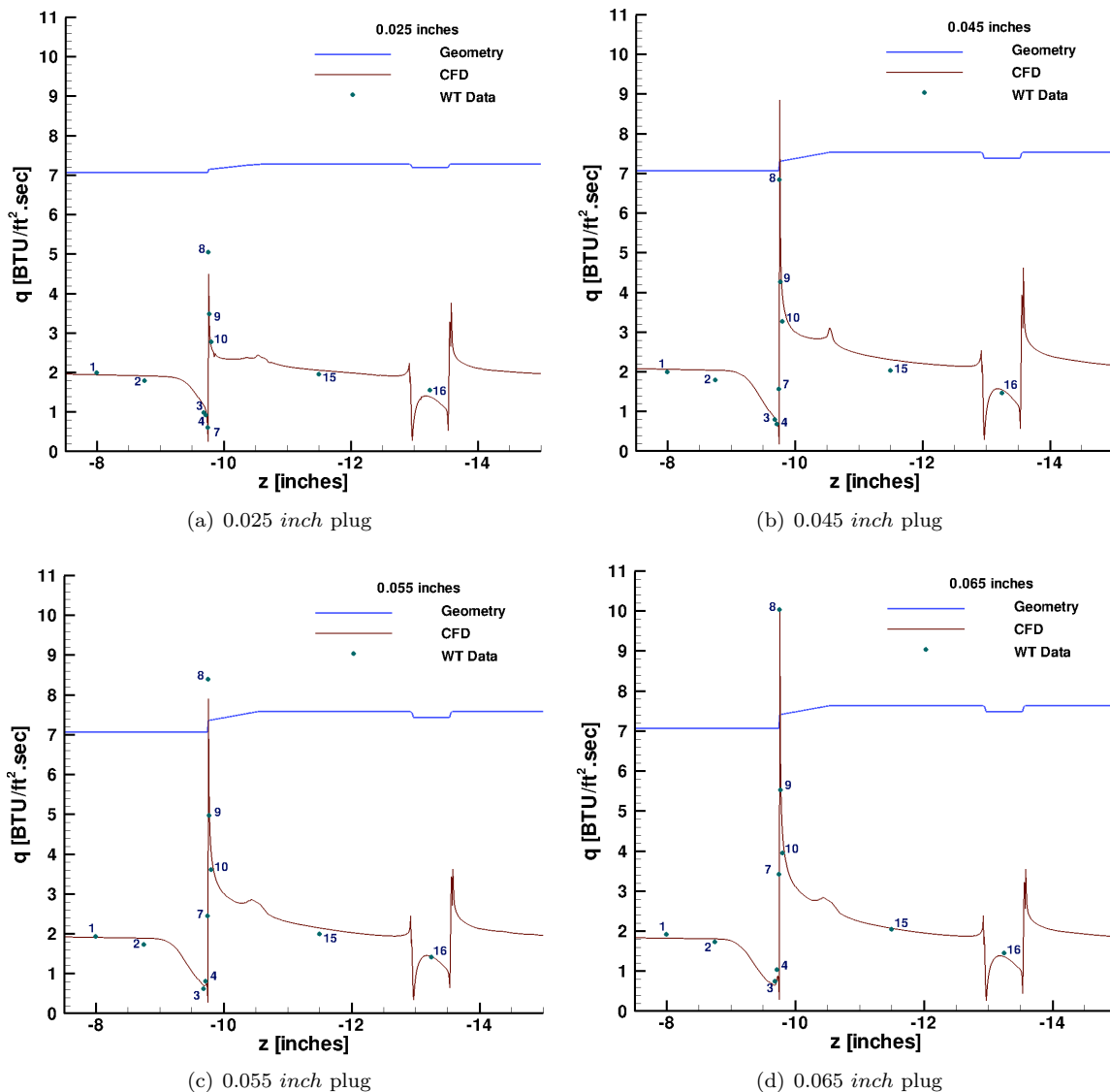


Figure 8. Heat flux variations along the Wind Tunnel (WT) geometry.

To compare heat fluxes on the WT model plug with different heights, the surface heat flux of each model is normalized by the local heat flux at probe location 1. This states the heating Bump Factor, BF. The results are plotted in Figure 9. The experimental BF data are also shown in this figure. As shown, BF on the plug top surface increases with plug height. To quantify the CFD accuracy, computed heat fluxes at the gauge locations 1-10 (see Figure 7) are plotted with the measured WT data in Figure 10 in which a good agreement is shown. In this figure, the WT errors are the repeatability errors. This figure shows that that the highest surface heat flux on the plug for all the cases occurs at gauge number 8. The difference between the highest computed and measured heat flux is within 5% for all the cases except for the 0.045 inch case in which it is 20%. As a result of the good comparisons, these data are employed to develop a generalized heating augmentation correlation that simplifies the prediction process. The intent is then to demonstrate the analysis can be formulated to flight conditions and can be used for flight analysis. This is described in the next section.

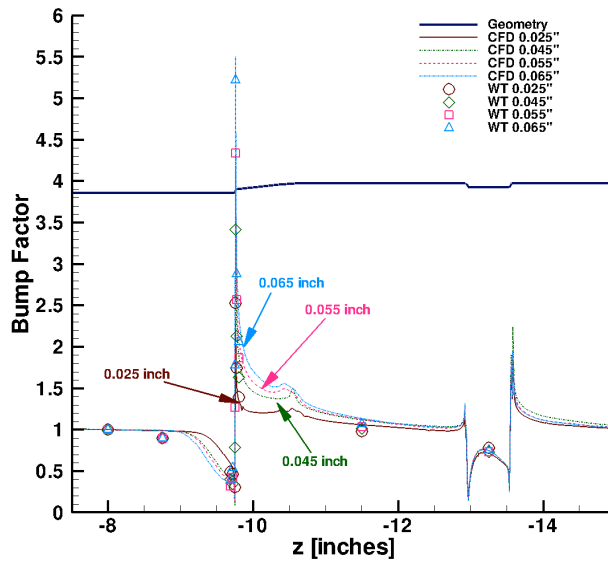


Figure 9. Heating Bump Factor, BF, along the Wind Tunnel (WT) geometry.

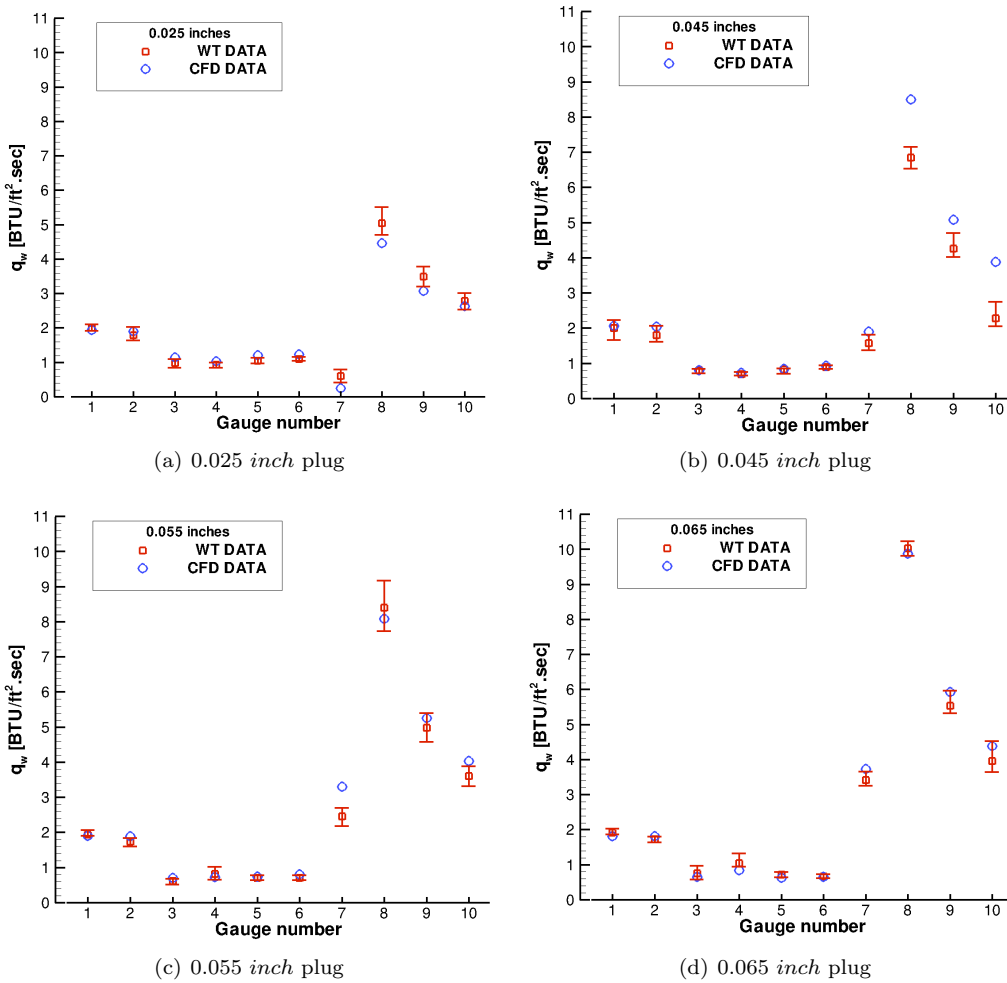


Figure 10. Wind Tunnel (WT) and CFD data comparison at the vicinity of the plug lip.

Heating Augmentation Correlation

The computational results and the WT data are used to develop a heating augmentation (HA) correlation for plug repair at flight conditions. This is obtained by calculating the peak heating rate at the lip of the plug, q_η , and the heat flux on the smooth surface without a plug at a location that would represent the base of the plug if it were present, $q_{base|no\ plug}$, (see Figure 11); i.e.

$$HA = \frac{q_\eta}{q_{base|no\ plug}}, \quad (2)$$

The measured peak heating distance above the surface is shown schematically in Figure 11 and tabulated in Table 4 for different plug heights. For the WT models, η is the location of the highest measured gauge.

To obtain heat fluxes at base of the plugs and on the smooth surface, q_{base} , simulations are performed on smooth WT swept cylinder geometries. The peak heating rates are calculated from the gauge 8 locations (see Figure 7.) These data are shown in Figure 12 in which the x-axis is the peak heating to free-stream enthalpy ratio, H_η/H_∞ . Heating augmentation on the Space Shuttle Orbiter WLE panels 9 and 18 (see Ref.¹²) are also computed and shown in this figure. The flight data show that a different correlation parameter is needed to correctly scale WT data (cold wall) to flight data (radiative equilibrium). LAURA is used for all CFD points.

An attempt was also made to examine k/δ as a correlation parameter. This parameter was used by Hung and Patel,¹¹ primarily, for large k/δ . They have shown a large uncertainty in heating augmentation associated with this parameter. Our collection of data, shown in Figure 13, also show that k/δ does not correlate well with wide range of conditions.

Examining the enthalpy profile, it was found that the enthalpy profiles of the WT and flight cases start from a different value on the surface due to different wall enthalpy values. Therefore, a correlation parameter was chosen that is based on the wall enthalpy values to collapse the WT and the CFD flight data; i.e. H_η/H_w . This correlation parameter used in Figure 14 in which the data are presented in log-linear format. Additional CFD flight cases from Ref.^{4,5} are also added in this plot. Protuberance to BL thickness ratio, k/δ , for all the presented cases are less than 0.3. It appears that all the presented data points but one are clustered around a single line.

The heating augmentation for the case where $H_\eta/H_w = 1$, which is 1, is also shown. This point is called theoretical point in this figure. The data are

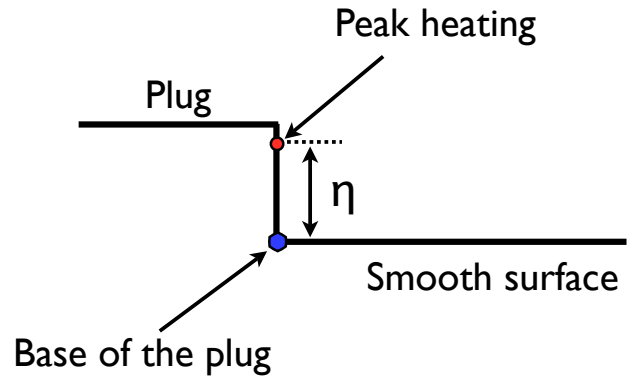


Figure 11. Schematic illustration for heating augmentation calculation (Eq. 2.)

Table 4. The distance from the base of the plug to a location where the peak heating is measured by gauge 8.

PLUG HEIGHT	η
<i>inches</i>	<i>inches</i>
0.025	0.0223
0.045	0.0423
0.055	0.0523
0.065	0.0623

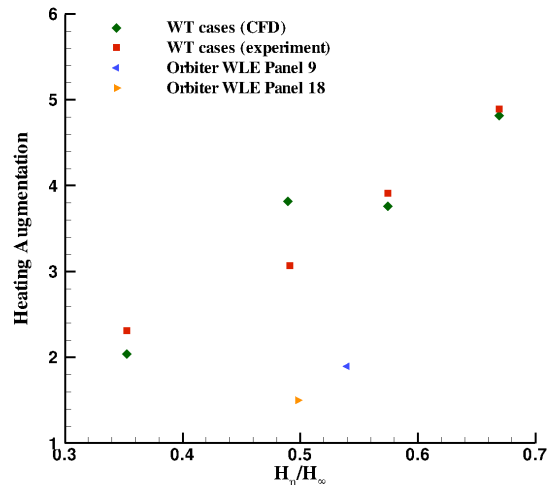


Figure 12. First attempt to scale WT to flight data.

curve-fitted by anchoring the data to the theoretical point, and the following correlation is obtained:

$$HA = 0.5 e^{0.7H_\eta/H_w} \quad (3)$$

Applying the Curve-Fit^a

To use the proposed heating augmentation correlation, only a flow field solution over an undisturbed Shuttle geometry is needed. The following steps are then needed to estimate the heating augmentation:

- locate smooth OML geometry that would represent the base of the protuberance if it were present^b
- obtain the wall enthalpy at the identified location, H_w , from the solution
- determine height above the surface that would represent height of the protuberance
- obtain the enthalpy at the identified protuberance lip location, H_η , from the undisturbed flow field solution
- calculate the enthalpy ration H_η/H_w
- use the proposed correlation equation, Equation 3, to estimate the heating augmentation.

Conclusions

Several CFD analyses are conducted and compared with WT results for a swept cylinder model with a protruding rounded plug. It is shown that the CFD data are in very good agreement with measured WT data. Based on the CFD and WT data, a heating augmentation correlation that is based on enthalpy profile is proposed. The proposed equation is examined for several CFD flight cases with short protuberances, where the protuberance to BL thickness ratio is less than 0.3, and a good agreement was achieved. Accordingly, the proposed correlation parameter shows potential for further investigation and testing against more flight cases.

Acknowledgment

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References

¹Columbia Accident Investigation Board Final report, Vols I-IV, URL: <http://caib.nasa.gov>, October, 2003.

^aThe curve-fit is only tested for small k/δ .

^bIt should be noted here that the base of the protuberance is also defined as the impingement point.

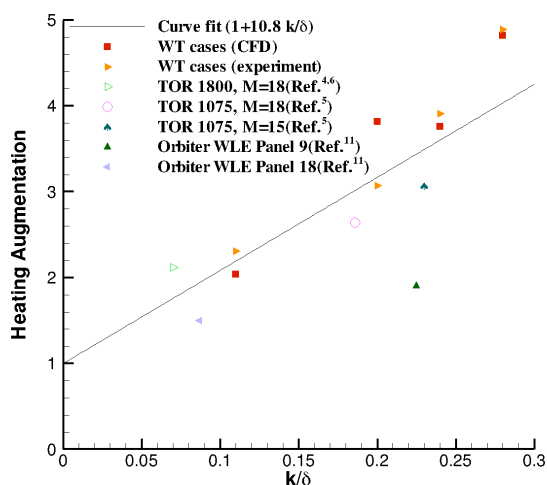


Figure 13. Heating augmentation based on k/δ parameter, which does not fit the data well.

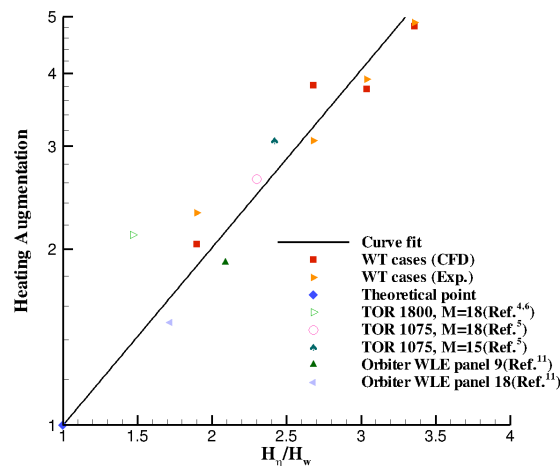


Figure 14. Proposed heating augmentation parameter for short hypersonic protuberances.

²Hung, F., and Patel, D., “Protuberance Interference Heating in High-speed Flow”, AIAA-84-1724, 1984.

³Lessard, V.R., “CFD-Predicted Tile Heating Bump Factors Due to Tile Overlay Repairs”, NASA Langley Research Center, NASA CR2006-214509, 2006.

⁴Mazaheri, A.R., “CFD Analysis of Tile-Repair Augers for the Shuttle Orbiter Re-entry Aeroheating”, NASA Langley Research Center, NASA CR2007-214858, 2007.

⁵Mazaheri, A.R., “Computational Aerothermodynamic Analysis of the Space Shuttle Orbiter Tile Overlay Repair with Different Geometries”, NASA Johnson Space Center, Aeroscience and Flight Mechanics Division, EG-SS-07-17, 2007.

⁶Mazaheri, A.R., and Wood, W.A., “Re-Entry Aeroheating Analysis of Tile-Repair Augers for the Shuttle Orbiter”, AIAA-2007-4148, 2007.

⁷White, T., and Tang, C., “CFD Support for CUBRC RCC Repair”, Aeroheating Panel Review Meeting, Johnson Space Station, Houston, TX, September 2007.

⁸Gnoffo, P.A., “An Upwind-Biased, Point-Implicit Relaxation Algorithm for Viscous, Compressible Perfect-Gas Flows”, NASA TP2953, February 1990.

⁹Gnoffo, P.A., Gupta, R.N., and Shinn, J.L., “Conservation Equations and Physical Models for Hypersonic Air Flows in Thermal and Chemical Nonequilibrium”, NASA TP2867, February 1989.

¹⁰“Post Test Report for the Aerothermal Wind Tunnel Verification Test OH-200 RCC repair C/SiC Plug and NOAX Crack Repair Model Configurations (Model 204-O)”, SE07HB008, The Boeing Company, 2007.

¹¹Hung, F., and Patel, D., “Protuberance Interference Heating in High-Speed Flow”, AIAA-84-1724.

¹²Mazaheri, A.R., “Computational Aerothermodynamic Analysis of Wing Leading Edge Plug Models with Comparison to Wind Tunnel Experiments”, NASA Johnson Space Center, Aeroscience and Flight Mechanics Division, EG-SS-07-18, 2007.

¹³Anderson, B., *Private Communication*, NASA Johnson Space Center, Houston, TX, 2007.