

# Confidence Testing of Shell 405 and S-405 Catalysts in a Monopropellant Hydrazine Thruster

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As part of the transfer of catalyst manufacturing technology from Shell Chemical Company (Shell 405 catalyst manufactured in Houston, Texas) to Aerojet (S-405 manufactured in Redmond, Washington), Aerojet demonstrated the equivalence of S-405 and Shell 405 at beginning of life. Some US aerospace users expressed a desire to conduct a preliminary confidence test to assess end-of-life characteristics for S-405. NASA Marshall Space Flight Center (MSFC) and Aerojet entered a contractual agreement in 2004 to conduct a confidence test using a pair of 0.2-lbf MR-103G monopropellant hydrazine thrusters, comparing S-405 and Shell 405 side by side. This paper summarizes the formulation of this test program, explains the test matrix, describes the progress of the test, and analyzes the test results. This paper also includes a discussion of the limitations of this test and the ramifications of the test results for assessing the need for future qualification testing in particular hydrazine thruster applications.

## I. Introduction

Following the Shell Corporation's decision to discontinue production of Shell 405 catalyst in 2002, aerospace users faced uncertainty regarding the continued availability of this alumina-based iridium catalyst which has for decades been used to decompose hydrazine in thrusters and auxiliary power units (APUs). With the ramifications of Shell's decision becoming clear, NASA's Marshall Space Flight Center (MSFC) and Kennedy Space Center (KSC) began work with Aerojet under contract number NAS8-02041 to assure timely and successful transfer of the catalyst manufacturing technology from Shell to Aerojet's Redmond, Washington facility under the new name, "S-405." Throughout this technology transfer effort, a group of NASA users convened regularly to monitor progress, discuss concerns, and evaluate potential issues in the technology transfer. On March 13, 2003, this group held an on-site meeting at Aerojet Redmond to review the outcome of the technology transfer and to close out the NASA contract that enabled it. At the conclusion of the meeting, representatives concurred that the manufacturing technology had been successfully transferred and that test results proved that 14-18 mesh S-405 and Shell 405 had equivalent beginning-of-life (BOL) characteristics.<sup>1</sup>

Some users, among them NASA MSFC propulsion engineering, began to express a desire for data demonstrating the equivalence of Shell 405 and S-405 near the end-of-life (EOL). At a July 24, 2003 meeting of the United States S-405 user community, MSFC set forth a preliminary concept for a confidence test to provide insight into EOL characteristics. The purpose of this test would be to identify a well-characterized thruster with a low thrust level (to minimize propellant costs) that had been previously qualified using Shell 405. Testing with S-405 would then provide a basis for comparison that could serve as an early exploratory test to uncover differences, if any, between the catalysts.

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Iterative discussions over the next several months refined the initial test approach, leading to user group consensus and a decision to proceed in January, 2004. This paper highlights the selected test approach and presents the results and conclusions of the test, which, due to various funding difficulties and schedule delays, began in January, 2005 and concluded two months later.

## II. Test Approach

The test approach continued to evolve for some time following the July, 2003 users meeting until Lockheed-Martin and RNR Engineering identified a pair of Aerojet MR-103G thrusters (see Figure 1) that would be available for continued testing after the conclusion of a Lockheed-Martin test in the same vacuum test position at Aerojet Redmond. Since the thrust level was low ( $0.2 \text{ lb}_f$ ) and since several thousand  $\text{lb}_f\text{-sec}$  of impulse would be accumulated before the confidence test would begin, the total propellant cost and total test cost was compatible with NASA's limited funding. Further, since the MR-103G had proven treacherous for Shell 405 in previous years, the group agreed that the test would be likely to uncover differences, if any, between Shell 405 and S-405 at EOL.

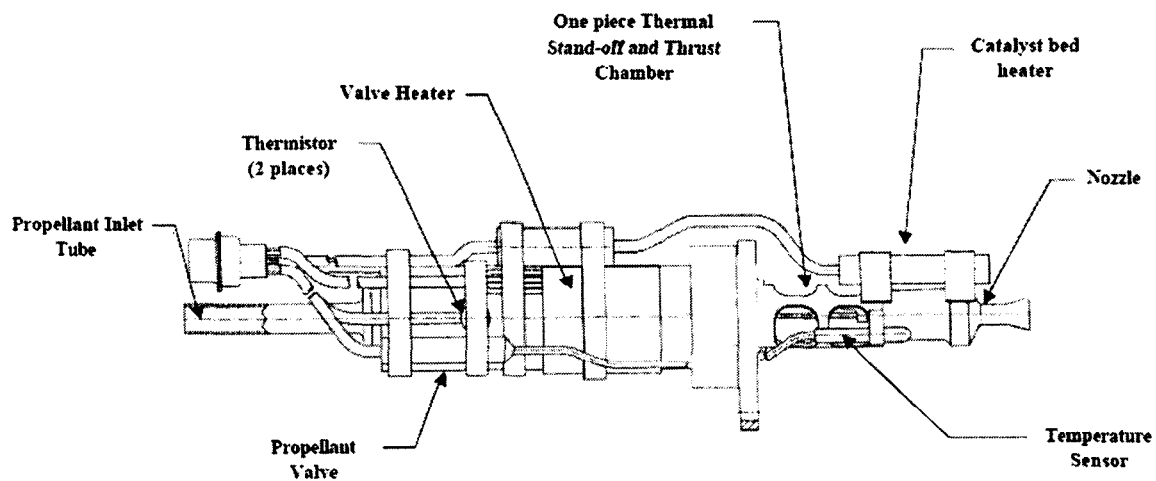


Figure 1. Depiction of MR-103G Thruster.

For clarity, the initial Lockheed-Martin test program will be referred to here as the "previous test program," whereas the subsequent testing will be called the "confidence test."

Figure 2 and Figure 3 illustrate the test conditions imposed in the previous test. As can be observed, the thruster burns were primarily long-duration (8-hour) continuous burns punctuated by periodic health checks.

Figures 4, 5, and 6 summarize the test conditions and flow for the confidence test. Pre-test evaluations included computer tomography (CT) scanning and functional checks (both electrical and mechanical). Unlike the previous test, the confidence test included a mixture of pulse mode operation (0.02 to 10 seconds "on") and periodic health checks. Following the side-by-side hot fire testing under vacuum conditions, post-test evaluations included the following: functional checks; CT scanning; disassembly and inspection; weighing and sieving the catalysts; Braunauer, Emmett, and Teller (BET) analysis; and hydrogen chemisorption testing.

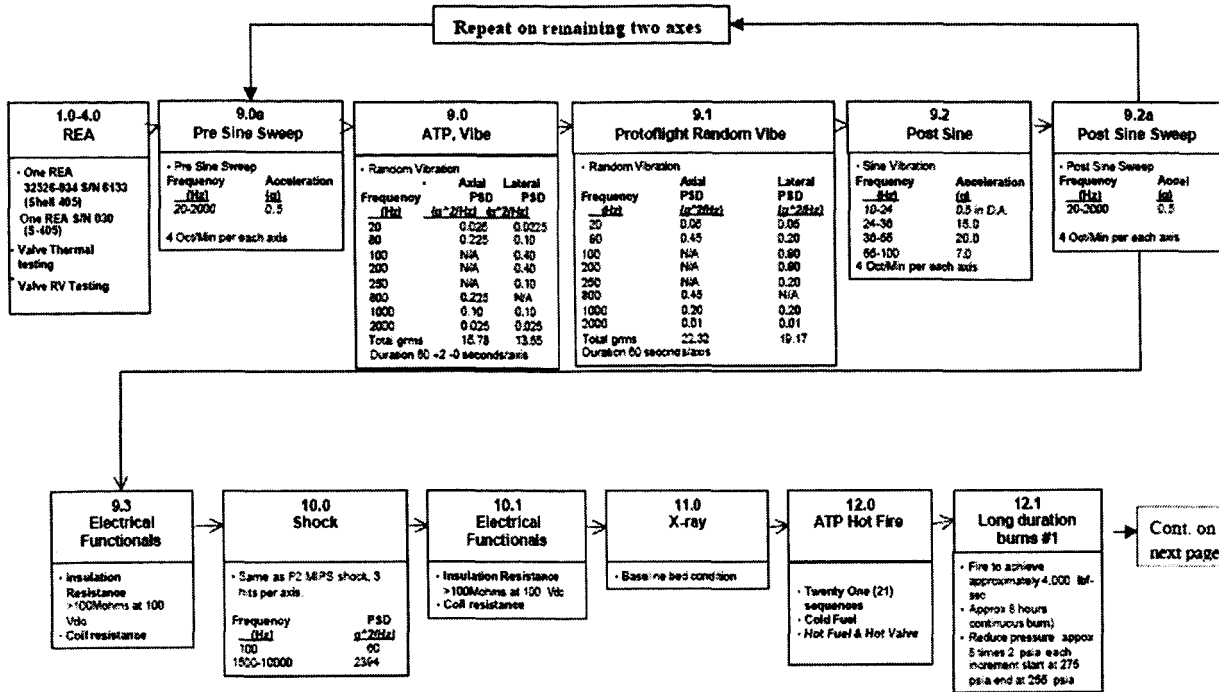


Figure 2. Previous Test Program Conditions and Flow (Part 1)

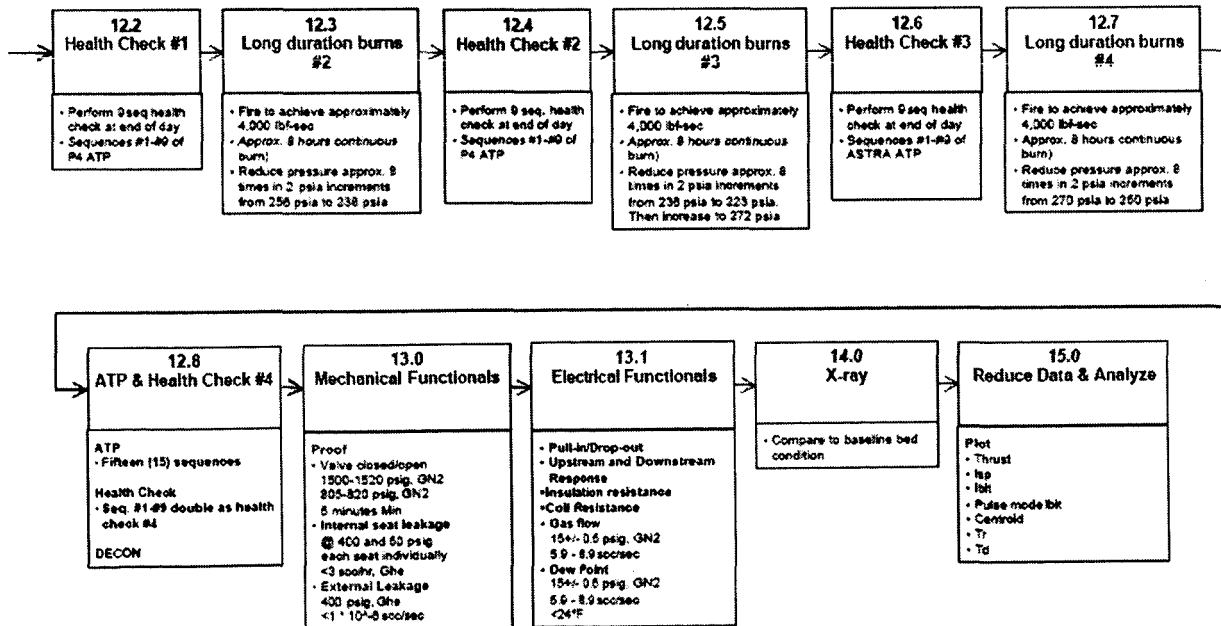


Figure 3. Previous Test Program Conditions and Flow (Part 2)

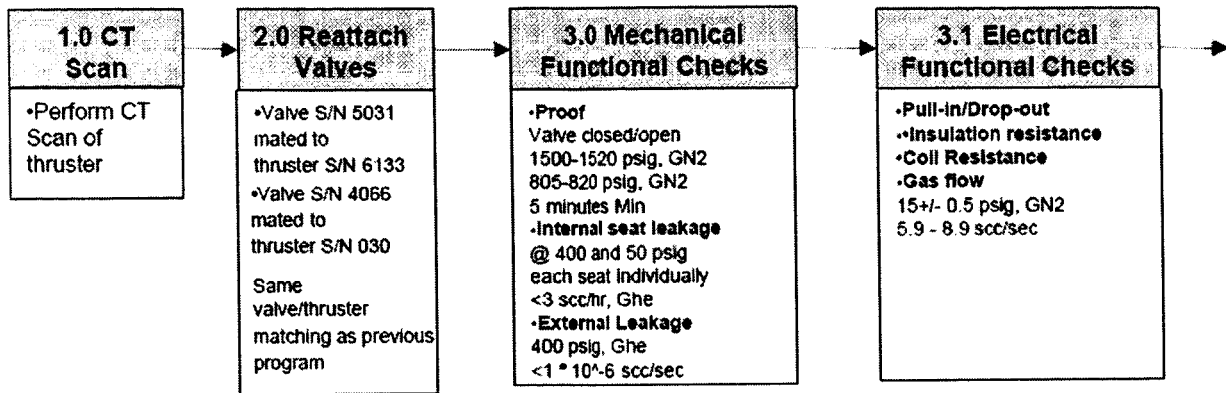


Figure 4. Test Conditions and Flow for the Confidence Test (Part 1)

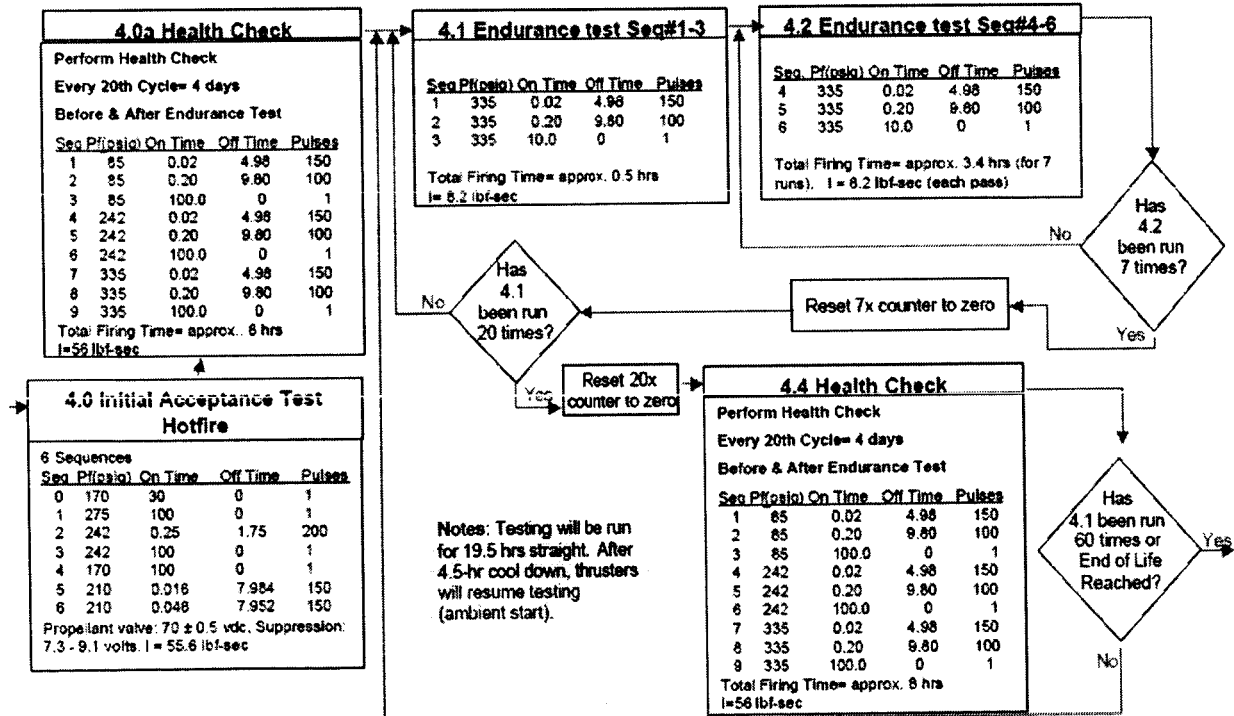


Figure 5. Test Conditions and Flow for the Confidence Test (Part 2)

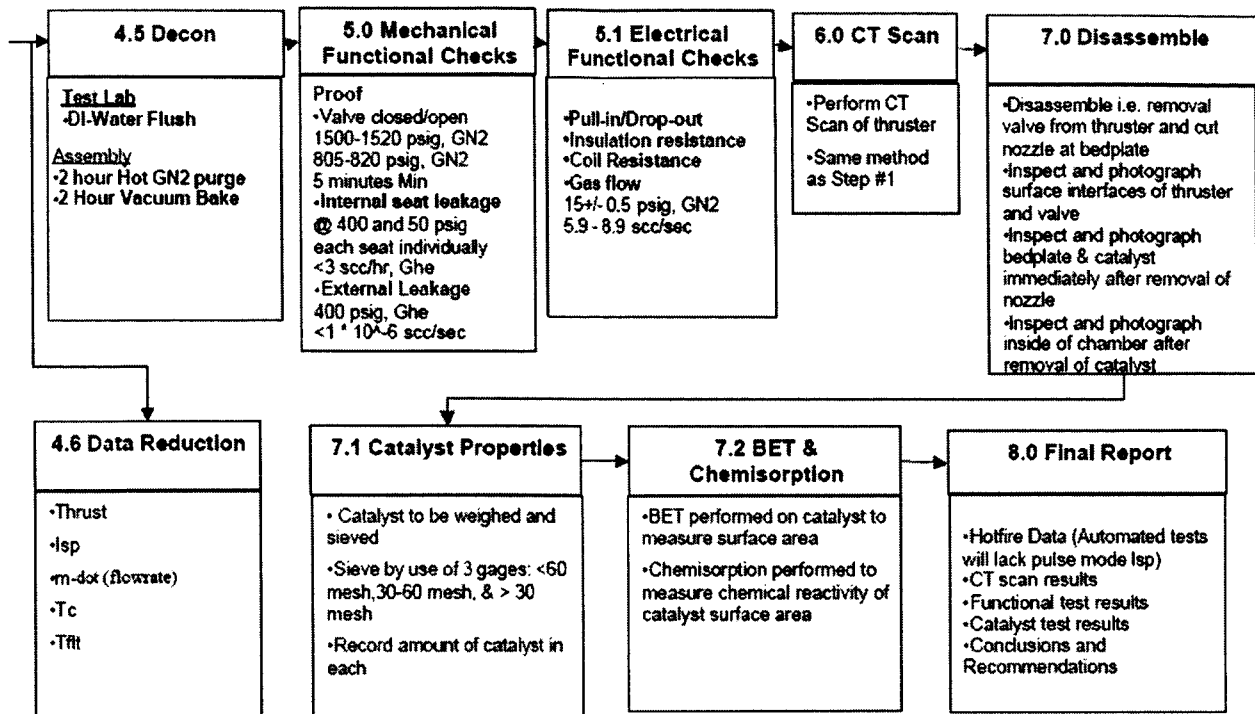


Figure 6. Test Conditions and Flow for the Confidence Test (Part 3)

### III. Test Results

The previous testing and confidence testing demonstrated the capabilities of the two MR-103G thrusters shown in

Table 1. Both thrusters successfully completed the hot fire test program, and all data was judged to be within historical parameters. Both thrusters intermittently experienced sining roughness, which is normal for the MR-103G thruster. In general, the S-405 thruster (S/N 030) appeared to sine more frequently. The health checks uncovered no unusual thermal operating conditions.

Evaluation of thrust vs. feed pressure and steady state specific impulse (Isp) vs. feed pressure indicated that both engines met BOL tolerances at both the beginning and end of the test. Steady state Isp was observed to dip below the minimum tolerance during the previous program's health checks. This behavior was attributed to the softness of the fuel system; the dissolved gas in the fuel affected both thrust and flow rate measurement.

Evaluation of thrust and Isp performance over the life of the thrusters indicated that performance differences were within typical engine-to-engine variation. At times, S-405 chamber pressure spiking was observed to a greater extent in the S-405 thruster (S/N 030).

In general there was no definitive pattern or trend observed apart from typical degradation associated with the accrual of life on the thrusters. Both thrusters experienced degradation in tail-off, due to formation of voids in the catalyst bed as life accrues. Data indicated no distinct trend in critical velocity (C-star). Also chamber pressure roughness had no pattern or trend observed apart from typical life degradation. Both steady state thrust levels and performance (Isp) levels were within BOL tolerances at the end of the test.

Pre-test CT scans of the Shell 405 thruster (S/N 6133) indicated a catalyst void of 0.4% of the total bed volume near the injector, while the post-test CT scan indicated voids totaling to 1.1% of the bed volume. On the other hand, the S-405 thruster (S/N 030) had no noticeable void in the pre-test CT scan, but had voids totaling 0.9% of the bed volume in the post-test scan. In general, the majority of post-test voids in both thrusters were near the injector. The scans were similar between the two thrusters, and the voids in both thrusters were judged to be in line with historical precedents.

The post-test disassembly and inspection revealed subtle differences in the condition of the two catalyst beds. The Shell 405 (S/N 6133) bed had a visibly observable void near the top of the catalyst bed, whereas the S-405 bed

had no observable void. When catalyst granules were poured out, the Shell 405 granules stuck together, indicating that sintering had occurred in the bed. S-405, however, poured freely (no sintering observed). Both thrusters exhibited typical chamber and bedplate discoloration and catalyst markings in the chamber.

BET testing indicated a higher BET value for S-405 than Shell 405 in the upstream portion of the beds. In the downstream portion of the beds, the opposite trend was observed. Post-test chemisorption analysis of both catalysts exceeded the typical value of 90  $\mu\text{mole H}_2/\text{g}$ .

**Table 1. Demonstrated Capabilities**

	Shell 405 Thruster (S/N 6133)	S-405 Thruster (S/N 030)
Total pulse count		
Previous Test	9,641	10,044
Confidence Test	124,010	124,010
Total	133,651	134,054
Number Of Ambient Starts	12	12
Cumulative Total Impulse ( $\text{lb}_f\text{-sec}$ )		
Previous Test	19,336	18,370
Confidence Test	3,841	3,812
Total	23,177	22,182
Longest Continuous Burn	8 hours 10 minutes <sup>a</sup>	8 hours 10 minutes <sup>a</sup>
Lowest Rate Duty Cycle Tested	0.16% (0.016 sec on, 9.984 sec off)	0.16% (0.016 sec on, 9.984 sec off)
Minimum Pulse Width (sec)	0.016 <sup>a</sup>	0.016 <sup>a</sup>
Inlet Pressure Range (psia)	100-350	100-350
Fuel Temperature	12-60°C <sup>a</sup> (54°F-140°F)	12-60°C <sup>a</sup> (54°F-140°F)
Valve Voltage Range (Vdc)	70-71 Seats wired in series	70-71 Seats wired in series
Valve Suppression Voltage (Vdc)	8.2	8.2
Change in Thrust @ 257 psia <sup>b,c</sup>	-3.6%	-4.3%
Change in $I_{sp}$ @ 257 psia <sup>b</sup>	-0.2%	-0.3%
Change in Roughness @ 257 psia <sup>b</sup>	+1.5%	+3.8%
%Catalyst Void (CT scan after all testing)	1.1%	0.9%
% Catalyst Void (By weight after all testing)	2.5%	0.6%
$\Delta\text{BET}$ , Upstream Bed ( $\text{m}^2/\text{g}$ )	-7	-8
$\Delta\text{BET}$ , Downstream ( $\text{m}^2/\text{g}$ )	-4	-8
$\Delta\text{Chemisorption}$ , Upstream Bed ( $\mu\text{moles H}_2/\text{g}$ )	-27	-26
$\Delta\text{Chemisorption}$ , Downstream Bed ( $\mu\text{moles H}_2/\text{g}$ )	-18	-17

<sup>a</sup> Indicates this was during the previous test program.

<sup>b</sup> The different changes in performance for the two engines are typical of engine-to-engine variation.

<sup>c</sup> 2-sigma roughness represents the range where 95% of the chamber pressure roughness is occurring (2 x standard deviation of  $P_c$  / Average  $P_c$ ). This better represents the average random roughness of the engines, whereas occasional spiking would bias peak-to-peak roughness and increase scatter.

#### IV. Conclusions

Based on these test results, the investigators have concluded that Shell 405 and S-405 catalysts provide equivalent performance near the end of life in an MR-103G 0.2- $\text{lb}_f$  thruster. Differences observed between these two thrusters are consisted with thruster-to-thruster variations.

It is appropriate to mention the limitations of this test program. While the user community generally concurred with the test approach reported here would be a useful exploratory test, there was also general consensus that no single test could assure S-405 equivalence for all thruster and APU applications. Hence, it must be emphasized that the intent of this test was to provide data that could be used as a basis for assessing confidence in S-405's EOL properties. While the positive test results may enhance confidence in the quality of S-405, MSFC does not intend for the test results to be used to certify S-405 as a "drop-in replacement" for Shell 405 for all applications and usage scenarios.

### **Acknowledgments**

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### **References**

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<sup>1</sup> Wucherer, E. J., Cook, T., Stiefel, M., Humphries, R., and Parker, J. "Hydrazine Catalyst Production: Sustaining S-405 Production Technology", AIAA 2003-5079, 39th AIAA/ASME/SAE/ASEE Joint Propulsion Conference & Exhibit, Huntsville, AL, July 20-23, 2003.