

Abstract

A key parameter in modeling differential spacecraft charging is the resistivity of insulating materials. This determines how charge will accumulate and redistribute across the spacecraft, as well as the time scale for charge transport and dissipation. Existing spacecraft charging guidelines recommend use of tests and imported resistivity data from handbooks that are based principally upon ASTM methods that are more applicable to classical ground conditions and designed for problems associated with power loss through the dielectric, than for how long charge can be stored on an insulator. These data have been found to underestimate charging effects by one to four orders of magnitude for spacecraft charging applications.

A review is presented of methods to measure the resistivity of highly insulating materials including the electrometer-resistance method, the electrometer-constant voltage method, the voltage rate-of-change method and the charge storage method. This is based on joint experimental studies conducted at NASA Jet Propulsion Laboratory and Utah State University to investigate the charge storage method and its relation to spacecraft charging. The different methods are found to be appropriate for different resistivity ranges and for different charging circumstances. A simple physics-based model of these methods allows separation of the polarization current and dark current components from long duration measurements of resistivity over day- to month-long time scales. Model parameters are directly related to the magnitude of charge transfer and storage and the rate of charge transport. The model largely explains the observed differences in resistivity found using the different methods and provides a framework for recommendations for the appropriate test method for spacecraft materials with different resistivities and applications. The proposed changes to the existing engineering guidelines are intended to provide design engineers more appropriate methods for consideration and measurements of resistivity for many typical spacecraft charging scenarios.

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 Title page Introduction Resistivity and spacecraft charging Definition of resistivity Thin Film Capacitor Model of Spacecraft Dielectrics Resistivity Charge Transport and Time Scales Time Scales and Spacecraft Charging Relevant scales for Spacecraft Charging 	 3. Two methods for measuring resistivity 1. Constant-voltage (ASTM) 1. Method 2. Instrumentation 3. Equations 4. Results for PET 2. Charge Storage Method 1. Method 2. Instrumentation 3. Equations 4. Results for: 1. PTFE
 Time Dependence of Capacitor Voltage: A Simple Model Capacitor voltage with leakage Capacitor voltage with polarization Polarization mechanisms and time scales Polarization in polymers A charge picture Two ways charge can change with time 	 FR4 Alumina Conclusions Time scales for measurements and polarization Instrument resolution Conclusions to be drawn

Resistivty and Spacecraft Charging As before. spacecraft accumulate Energetic Magnetospheric Ions & Electrons charge and adopt potentials in Backscattered response to the plasma environment. Electrons Souttered Ions Secondary Jectrons (SE's) Ambient The distribution and migration of this charge determines the extent of differential charging. Resistivity of insulating materials -emitted determines: Spacecraft Charging results from Where charge will charge accumulation. accumulate How charge will redistribute across the spacecraft • Time scale for charge transport and dissipation.

8-April-05

9th SCTC 2005

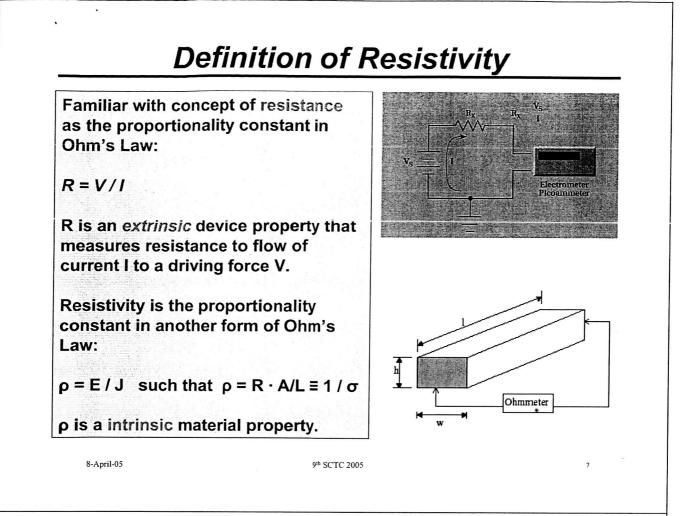
Our Spacecraft Charging Issues

New testing have identified a problem

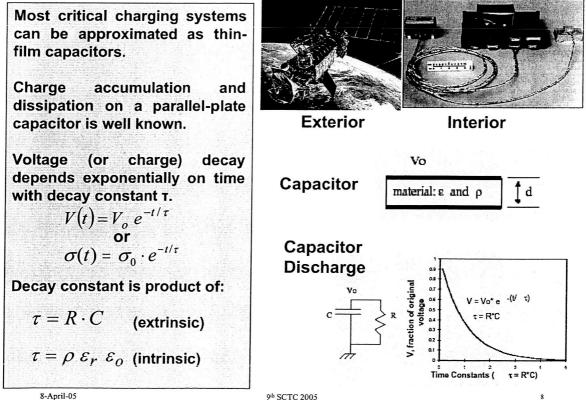
- Charge Storage resistivity tests done on Polyimides, Mylar, Teflon, Glass, Circuit Boards, etc. (see Green).
- Results from new resistivity methods find p 10¹-10⁴ times larger than handbook ASTM values.
- Charge can accumulate from many orbits.

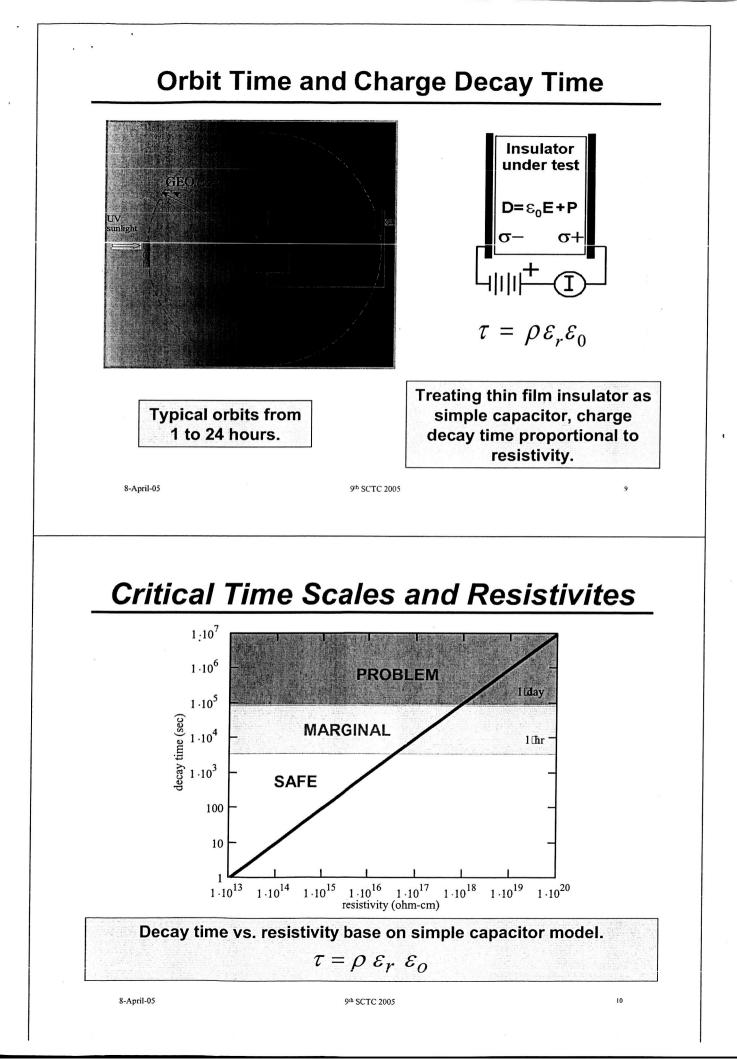
What voltages/charge distributions are developed?

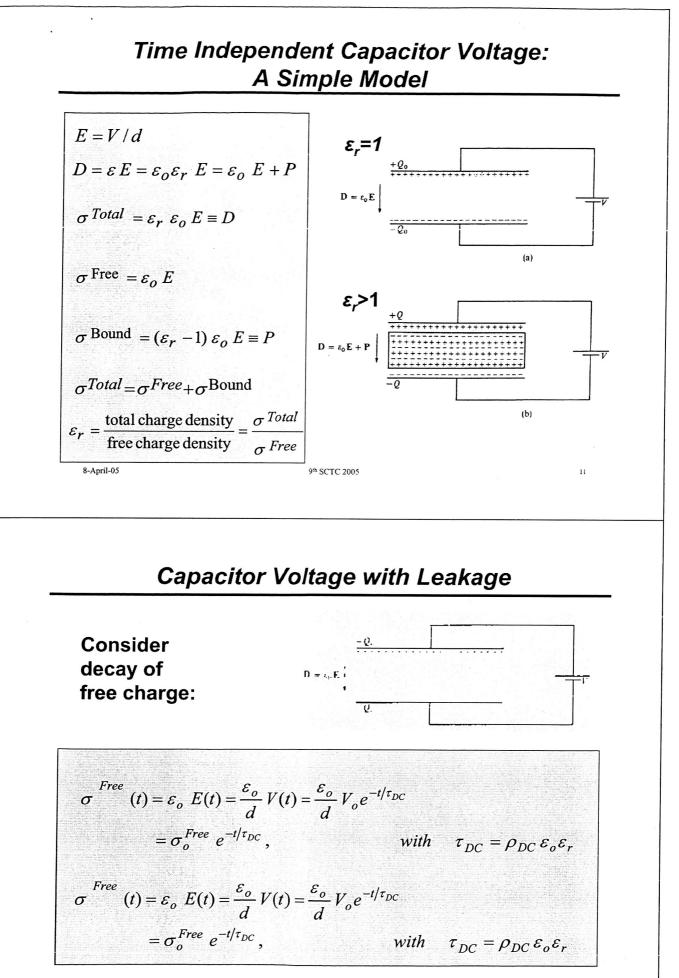
- > What are the proper test procedures?
- How do we qualify a material for space flight?
- What are mechanisms of charge storage and dissipation?

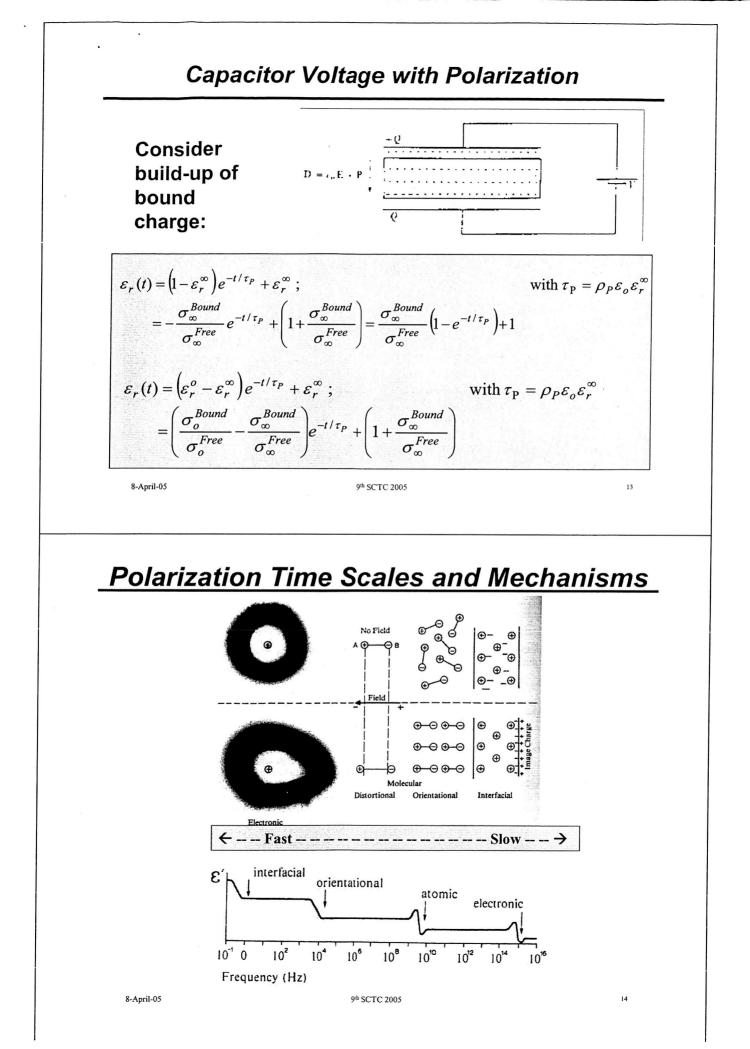


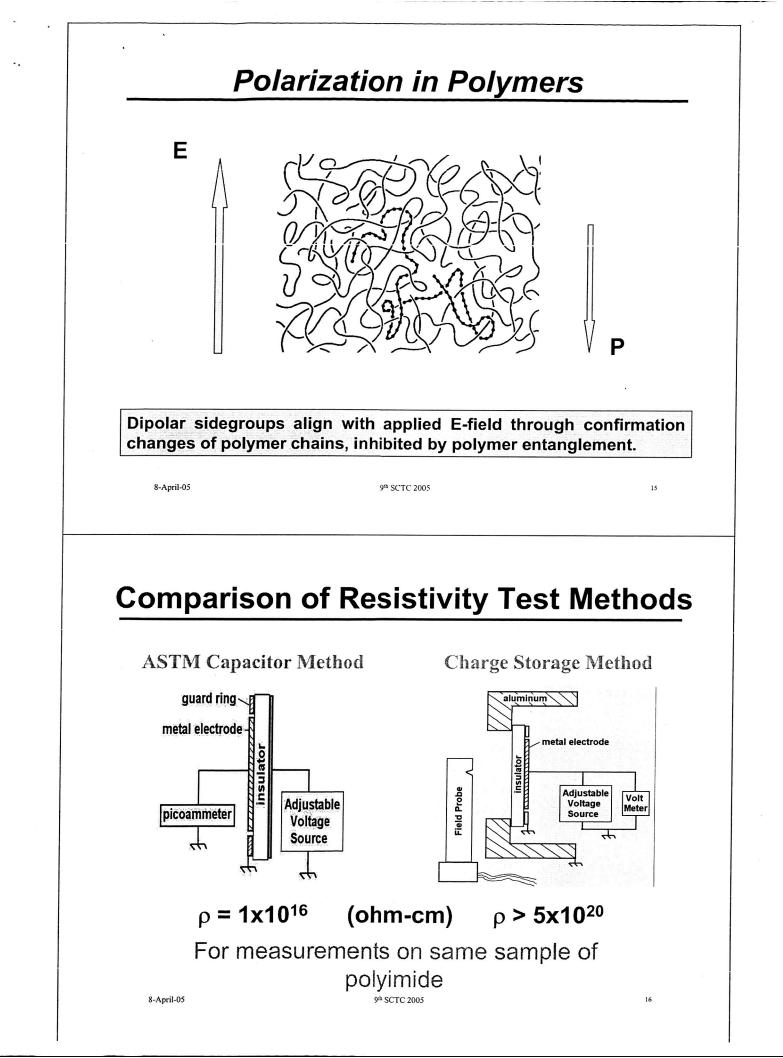
Thin-film Capacitor Model for Spacecraft Dielectrics

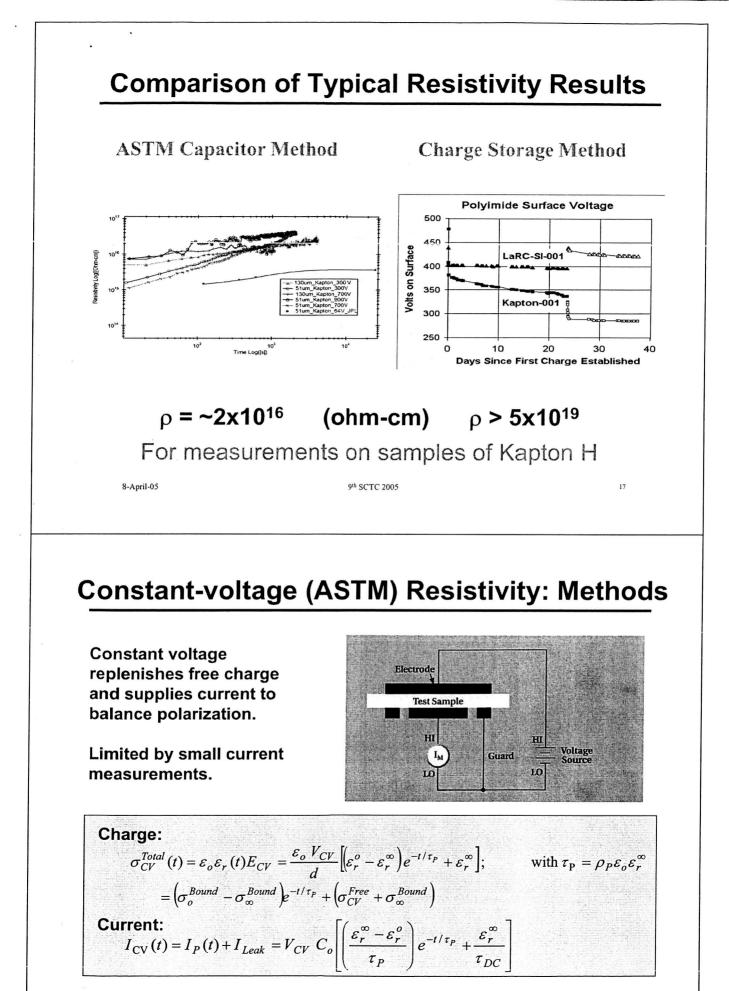






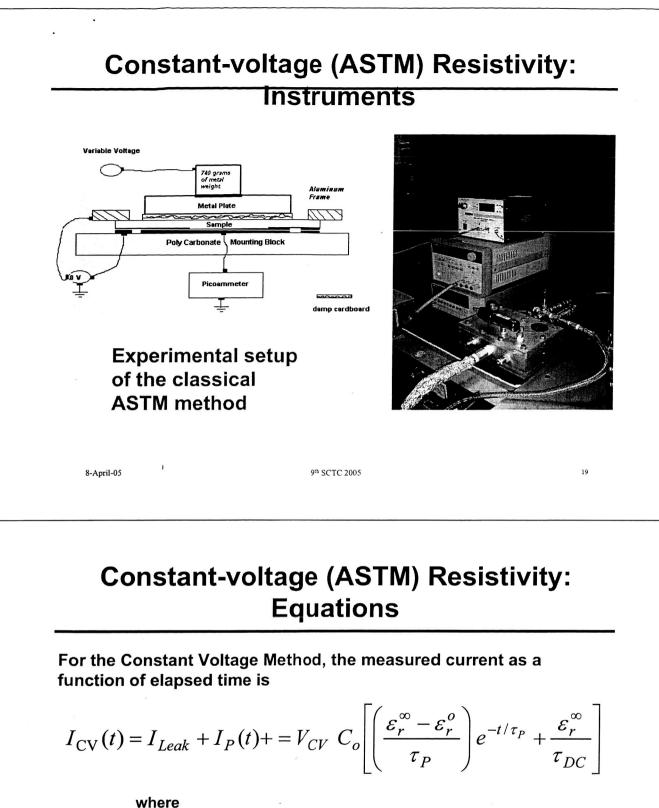




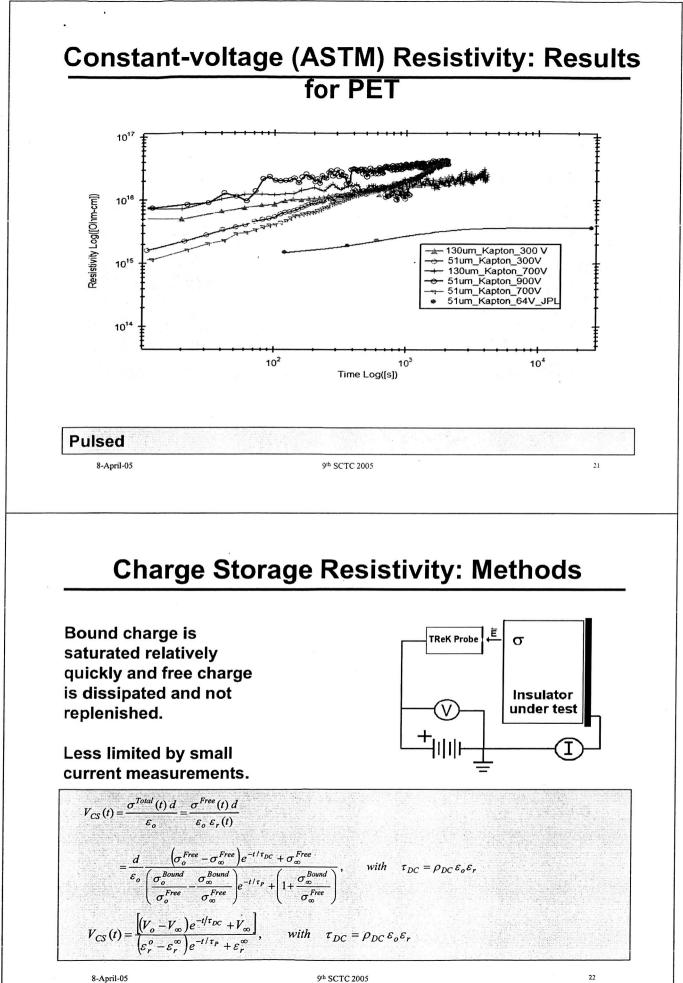


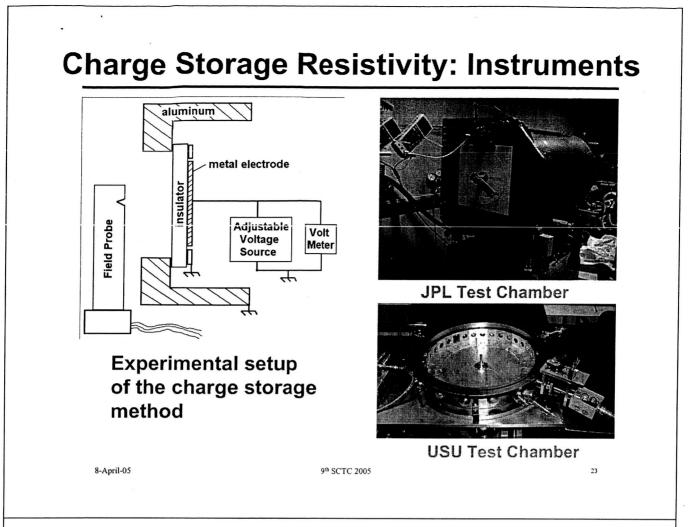
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t = time, seconds t_p = polarization decay constant, seconds t_{DC} = dark current decay constant, seconds e_r^{o} = initial relative dielectric constant, F/m e_r^{∞} = asmyptotic relative dielectric constant, F/m l_{CV} =measured current, amp V_{CV} =constant applied voltage, volt C_o = capacitance of the sample with e=1, farads





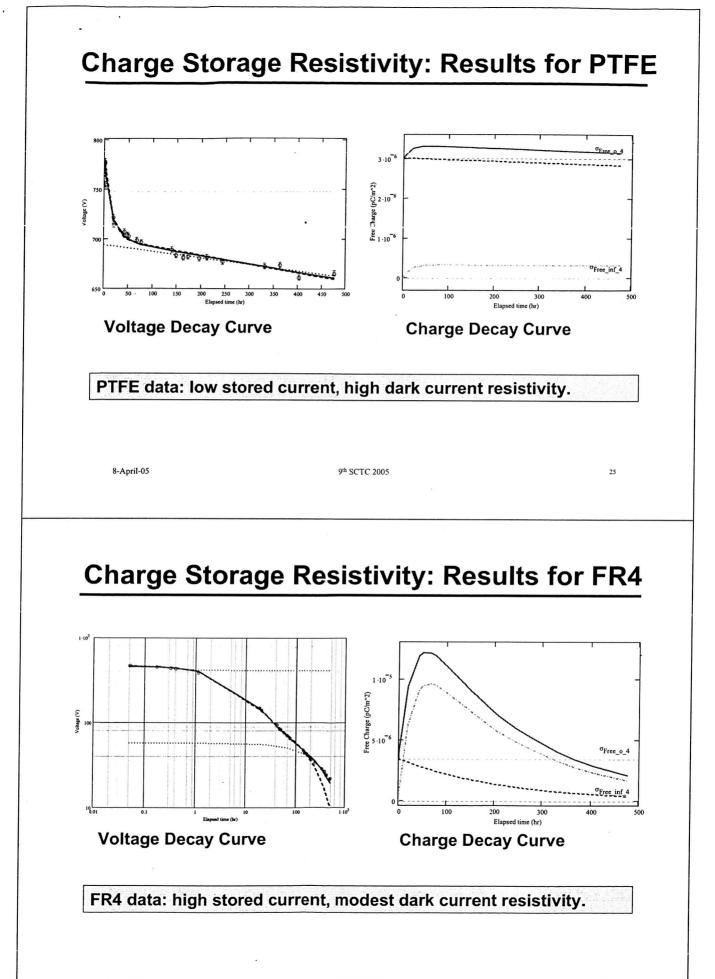
Charge Storage Resistivity: Equations

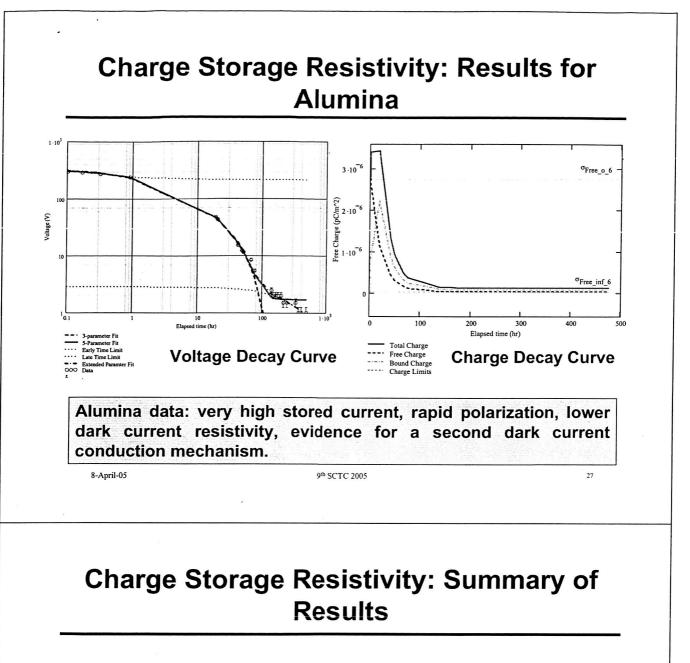
For the Charge Storage Method, the measured current as a function of elapsed time is:

$$V_{CS}(t) = \frac{\left[\left(V_o - V_\infty \right) e^{-t/\tau_{DC}} + V_\infty \right]}{\left(\varepsilon_r^o - \varepsilon_r^\infty \right) e^{-t/\tau_P} + \varepsilon_r^\infty},$$

with $\tau_{DC} = \rho_{DC} \varepsilon_o \varepsilon_r$

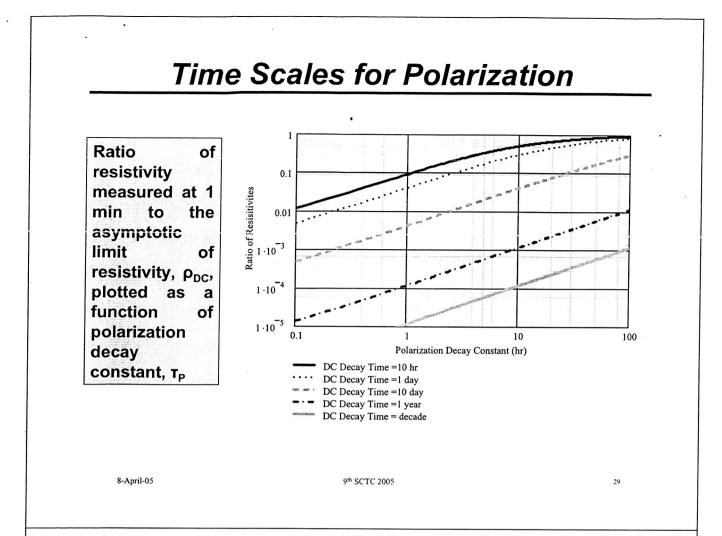
where t = time, seconds t_p = polarization decay constant, seconds t_{DC} = dark current decay constant, seconds e_r^{o} = initial relative dielectric constant, F/m e_r^{∞} = asymptotic relative dielectric constant, F/m Vo= initial voltage, volts V_{∞} = asymptotic voltage, volt C_o = capacitance of the sample with e=1, farads





Experimentally Determined Resistivity values for CRRES IDM samples

Material	Thickness	εr°	εr∞	V_o	V_{∞}	τ_{P}	τ_{DC}	ρ _{5 parameter}	P _{3 paramter}	ρ _{5 parameter}
-	(cm)			(volt)	(volt)	(hr)	(day)	(Ω-cm)	$(\dot{\Omega}-cm)$	/PASTM
PTFE	0.229	1.05	1.11	347	5.2	17.9	339	3.0×10 ²⁰	2.9×10 ²⁰	3×10 ²
FR4	0.317	1.07	1.95	412	1.5	18.2	4.53	2.3×10 ¹⁸	2.1×10 ¹⁸	<2×10 ⁹
Alumina	0.102	1.02	3.00	423	4.7	6.35	21.3	2.9×10 ¹⁷	3.0×10 ¹⁷	3×10 ³



Instrument Resolution

Method	Maximum Detectable Resistance Values and	Typical Maximum Measurable Values (±6%)						
	Decay Time Constant ^e	Resistance	Current	Resistivity	Decay Time Constant ^d			
Digital Multimeter	$ \begin{array}{l} \sim 2 \cdot 10^{10} \Omega / \\ \sim 5 \text{ sec}^{b,d} \end{array} $	~10 ¹⁰ Ω	~5·10 ⁻⁹ A	$\sim 1.10^{12} \Omega \cdot cm$	0.1 sec			
Electrometer— Resistance	$\sim 10^{16} \Omega /$ $\sim 3 \text{ days}^{b,d}$	~10 ¹⁴ Ω °	~5·10 ⁻¹² A	$\sim 1.10^{16} \Omega.cm$	<45 min			
Electrometer— Constant V	~5·10 ¹⁷ Ω / ~150 days °	~5·10 ¹⁶ Ω	~1·10 ⁻¹³ A ^{-b,d}	$\sim 5 \cdot 10^{17} \Omega \cdot \mathrm{cm}$	<1.5 day			
Voltage Rate- of-change	$\sim 4.10^{18} \Omega / \sim 3 \text{ yr}$ (R _{max} C=10 ⁸ $\Omega \cdot \text{F}$) ^{b,e}	$-4 \cdot 10^{16} \Omega$ (R _{max} C=10 ⁶ $\Omega \cdot F$) ^c	~1·10 ⁻¹⁴ A	~4·10 ¹⁸ Ω·cm	<12 day			
Charge Storage Decay	$\frac{-1 \cdot 10^{20} \Omega}{(R_{max}C=2.10^{2} \Omega)^{g}} \frac{1}{(2.10^{2} \Omega)^{g}}$	$-2 \cdot 10^{19} \Omega$ (R _{max} C=4 \cdot 10 ⁸ Ω · F)		$-2.10^{21} \Omega \cdot cm$.	<15 yr			

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Summary for Resistivity Test Methods Model

> Instrumentation and methods have been successfully developed to measure resistivity with charge storage decay method and compare the results with classical method.

Measurements confirm initial results that charge storage resistivity can be >10⁴ times classical results.

> Theoretical model based on simple physical parameters:

> Fits time-dependant data from different methods

Predicts disparities between different methods

> Explains resolution limits of different methods

Confirms charge storage method as method of choice for very high resistance materials

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