Innovative Mars Global International Exploration (IMaGInE) Mission

Davide Conte
Ph.D. Candidate, The Pennsylvania State University, USA

Dorota Budzyń
M.S. Student, Wrocław Institute of Technology, Poland

Hayden Burgoyne
Ph.D. Candidate, California Institute of Technology, USA

Marilena Di Carlo
Ph.D. Candidate, University of Strathclyde, UK

Dan Fries
Ph.D. Candidate, Georgia Institute of Technology, USA

Maria Grulich
M.S. Student, Technische Universität München, Germany

Sören Heizmann
M.S. Student, Universität Stuttgart, Germany

Henna Jethani
M.S. Student, University of Colorado Boulder, USA

Mathieu Lapôtre
Ph.D. Candidate, California Institute of Technology, USA

Tobias Roos
Ph.D. Candidate, Luleå University of Technology, Sweden

Encarnación Serrano Castillo
Ph.D. Candidate, Università di Bologna, Italy

Marcel Scherrmann
M.S. Student, Universität Stuttgart, Germany

Rhiannon Vieceli
M.S. Student, The Pennsylvania State University, USA

Lee Wilson,
Ph.D. Candidate, California Institute of Technology, USA

Christopher Wynard
B.S. Student, University of Illinois Urbana-Champaign, USA

This paper presents the conceptual design of the IMaGInE (Innovative Mars Global International Exploration) Mission whose mission objectives are to deliver a crew of four astronauts to the surface of Deimos and a robotic exploration mission to Phobos for approximately 343 days during the years 2031 and 2032, perform surface excursions, technology demonstrations, and In Situ Resource Utilization (ISRU) of the Martian moons as well as site reconnaissance for future human exploration of Mars. This is the winning mission design of the 2016 Revolutionary Aerospace Systems Concepts - Academic Linkage (RASC-AL) competition, awarded with the "Best in Theme," "Best Overall," and "Pioneering Exceptional Achievement Concept Honor (PEACH)" prizes. This competition was sponsored by NIA and NASA.

I. Introduction

Space exploration enriches and strengthens humanity's future by bringing nations together for a common cause; it reveals knowledge, inspires and educates people, creates a global partnership and establishes a sustained human presence in the Solar System by extending human frontiers and stimulating technical and commercial innovation on Earth. Sustainable space exploration is a challenge that no one nation can do on its own. To this aim, the Global Exploration Strategy, which was agreed on and published in May 2007 by fourteen space agencies, reflects a determination to explore our nearest neighbors: the Moon, asteroids, and Mars. In this framework, the Dream Team has been created with young engineering and applied science students from all over the world with a common goal, the IMaGInE Mission.

The **I**nnovative **Ma**rs **G**lobal **In**ternational **E**xploration (IMaGInE) Mission is the resulting work of 15 students gathered from 11 universities and 8 different nations spanning 6 different time zones with the objective of creating a mission design for the Crewed Mars Moons Mission theme proposed by the 2016 Revolutionary Aerospace Systems Concepts - Academic Linkage (RASC-AL) competition sponsored by NIA and NASA. In June 2016, the IMaGInE Mission was presented at the 2016 RASC-AL Forum in Cocoa Beach, FL and it was awarded with the "Best in Theme," "Best Overall," and "Pioneering Exceptional Achievement Concept Honor (PEACH)" prizes. The student names are listed as authors starting with the team leader (Davide Conte) and then listed alphabetically for the remaining 14 students.

II. Mission Architecture and Test Mission

The IMaGInE mission will deliver a crew of four astronauts to the surface of Deimos and a robotic exploration mission to Phobos for approximately 343 days during the years 2031 and 2032. The crew will perform surface excursions, technology demonstrations, and In Situ Resource Utilization (ISRU) of the Martian moons as well as site reconnaissance for future human exploration of Mars. The IMaGInE Mission is divided into two main segments: the test mission and the main mission. The test mission first provides the opportunity to test all of the major subsystems combined together in space, thus raising the overall system's Technology Readiness Level (TRL). Additionally, the test mission substantially lowers the risk the main mission crew incurs and leaves the science portion of the mission untouched. A summary of IMaGInE's mission architecture is depicted in Figure 1. This diagram also shows when and where supplies are replenished (REV-1, REV-2, REV-3, REV-4). The mission architecture is explained in detail in the following paragraph.

The first launch takes the Martian Moons Resupply and Science Deployment (MMRSD) vehicle into Low Earth Orbit (LEO) in January 2025. This launch is performed using a NASA Space Launch System (SLS) Block 1B from the Kennedy Space Center (KSC) and consists of a Resupply Expendable Vehicle (REV) that is pre-deployed at Deimos to ensure that the crew has enough supplies to conduct scientific exploration of the Martian system (Mars, Phobos, and Deimos). Along with resupply vehicle REV-4, a science payload is to be delivered at Phobos and Deimos. More details about the scientific part of the mission can be found in the *Science and Robotics* section. MMRSD consists of an Asteroid Redirect Mission (ARM)-derived propulsion system with a Multi-Purpose Logistics Module (MPLM)-derived module (REV-4) containing supplies for the crew. The spacecraft performs a low-thrust interplanetary transfer (Table 14 in Appendix F) and arrives in an orbit similar to that of Deimos in early April 2030. Note that although MMRSD is launched relatively early compared to the other launches, it reuses technologies that would be available for ARM in the early 2020s.

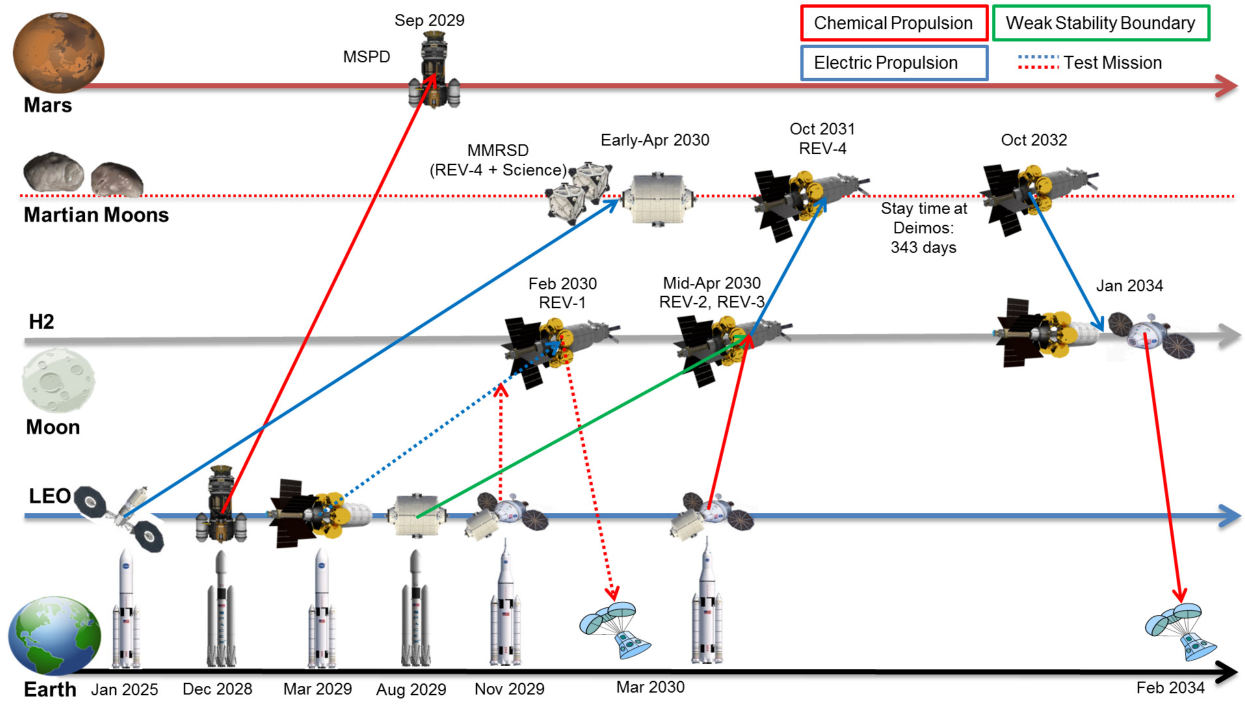


Figure 1. IMaGInE's mission architecture.

In December 2028, a Falcon Heavy is launched from KSC carrying scientific instruments that are delivered to the Martian surface, the Mars Surface Payload Deployment (MSPD), arriving in September 2029 via an interplanetary Hohmann transfer. In the meantime, the test mission begins with an uncrewed SLS Block 1B which launches from KSC in March 2029. This launch takes HERMES (Human Electric Reusable Mars Earth Shuttle), which houses the primary propulsion and power systems of the mothership, and HARMONIA (Habitable Ark for Mars Operations and Interplanetary Activities), the habitat used by the crew during the mission, into LEO. From LEO, the mothership (HERMES + HARMONIA) begins a low-thrust maneuver. A test crew is launched on top of an SLS Block 1B in early November 2029 so that they can arrive at the mothership once both spacecraft reach an altitude of approximately 60,000 km from Earth's surface in mid-November 2029. This altitude was chosen to perform the rendezvous of the two spacecraft because it minimizes the time the test crew spends in the Van Allen Belts radiation region. While the mothership takes 252 days to arrive at 60,000 km, the crew uses Orion's main engine to arrive at the same location in about 10 days. The test crew launch consists of a crewed Orion capsule and a resupply module, REV-1, that carries resupplies for the mothership for the test mission (Figure 2).

Once the test crew arrives at the mothership and the resupply has been completed, REV-1 is discarded and the mothership + test crew in Orion continue to spiral out via a low-thrust maneuver until they reach the Earth-Moon Lagrange Point 2 (EML2). Here, the spacecraft completes an insertion maneuver into a halo orbit about EML2 (H2) in February 2030. At this point the test crew undocks from the mothership and performs a lunar flyby to return to Earth in approximately 10 days. As the test mission ends, the ground crew is given system performance and systems-crew interaction data from which it can be decided if the main mission can be carried out. The main mission begins in March 2030, when a new crew launches on board of Orion with an SLS Block 1B from KSC, bringing a second resupply spacecraft, REV-2, capable of resupplying the mothership in a similar way done by the test crew (Figure 2), this time at H2. A third resupply mission (REV-3), which is delivered by a Falcon Heavy on a Weak Stability Boundary (WSB) trajectory, arrives and prepares the mothership for the journey to Deimos (resupply procedure shown in Figure 3). In mid-April 2030, the mothership + Orion depart H2, performing an interplanetary low-thrust maneuver, and arrive in the Martian Sphere of Influence (SOI) in late August 2031. The spacecraft arrives at Deimos in October 2031 where the crew performs the fourth resupply mission (REV-4) which was pre-deployed by MMRSD (resupply procedure shown in Figure 3). Once the resupply takes place, scientific

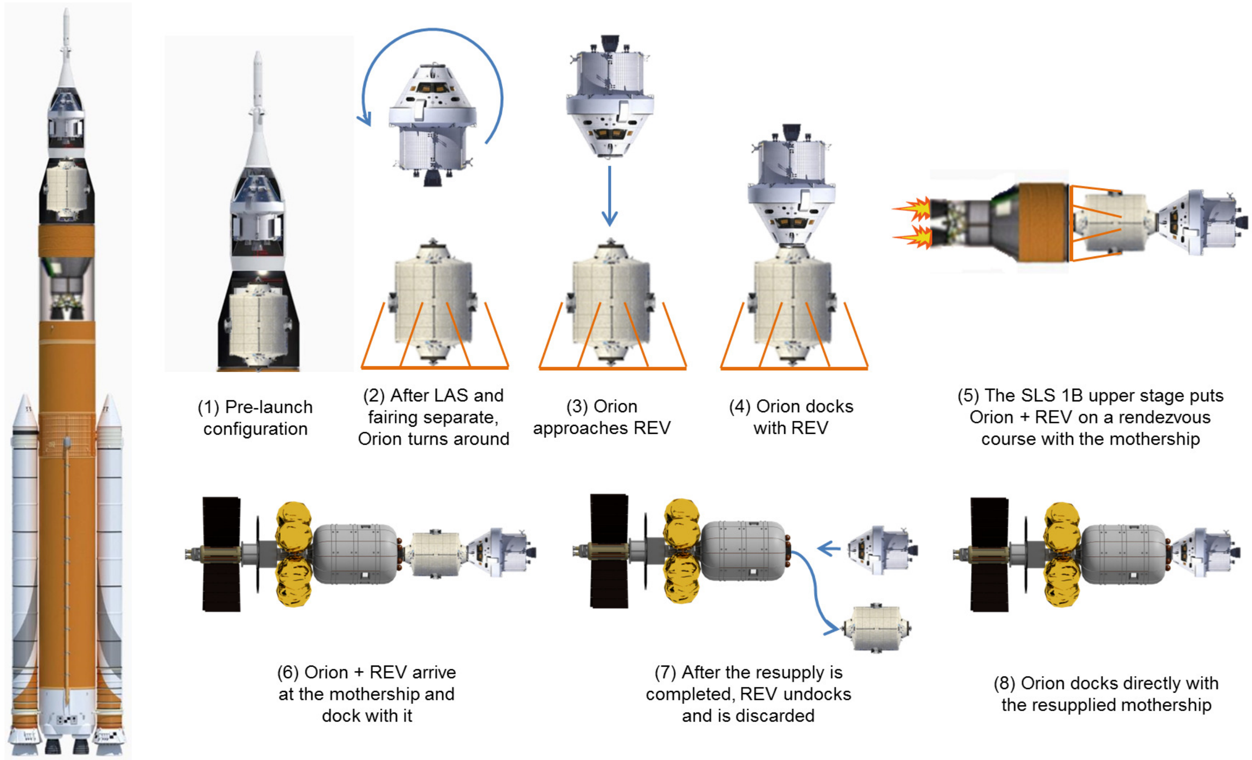


Figure 2. Main phases of the first two Resupply Expendable Vehicles, REV-1 and REV-2.

operations ensue for approximately 340 days. In October 2032, the crew departs from Deimos and returns to Earth's SOI in January 2034. Upon arrival in Earth's SOI, the crew separates on board Orion and performs a direct re-entry, while in late January 2034 the mothership returns to H2 for future resupply and reuse. A computer-generated model of the entire spacecraft is visible in the attached Compliance Matrix. For a short video of the mission see the mission video.[1]

Note that each REV is fitted with two docking ports located on opposite ends of the vehicle so that one docks with the mothership and the other docks with Orion. Having two docking ports on each REV avoids having to depressurize and re-pressurize Orion. The resupply procedure utilized by REV-1 and REV-2 is shown in Figure 2 while that used by REV-3 and REV-4 is shown in Figure 3. Additionally, REV-1, REV-2, and REV-4 are MPLM-derived spacecraft while REV-3 consists of a smaller ATV-derived module.

III. Mission Analysis

In order to accomplish the mission, the mothership's main propulsion system is a series of four Variable Specific Impulse Magnetoplasma Rockets (VASIMR) which are powered by a series of Safe Affordable Fission Engines (SAFE-400)[2].[3] In order to shield the crew from the SAFE-400s on board, additional reactor shielding based on the X-ray telescope Chandra is used. This is composed of slightly curved mirrors that are used to diffract X-rays away from HARMONIA.[4] Compared to chemical and nuclear propulsion, using electric propulsion reduces the required Initial Mass in LEO (IMLEO) for round trips to Mars by at least one order of magnitude. Chemical propulsion is only used to reduce the Time of Flight (ToF) of the crew from LEO to H2 at departure and from H2 to LEO at arrival. IMaGInE's architecture is developed with the idea of making missions to the Martian system sustainable and cost-efficient. In fact, the mothership is kept in H2 for future missions. H2 was chosen as the staging location for the mission because it allows constant communication and is a favorable energetic orbit close to Earth, from which the crew can return to Earth and to which the crew can easily arrive in at most 10 days using chemical propulsion. Figures 4 and 5 show the crewed interplanetary outbound and inbound trajectories of IMaGInE where green and red symbolize coasting and thrusting, respectively. Details regarding the method adopted for computing such

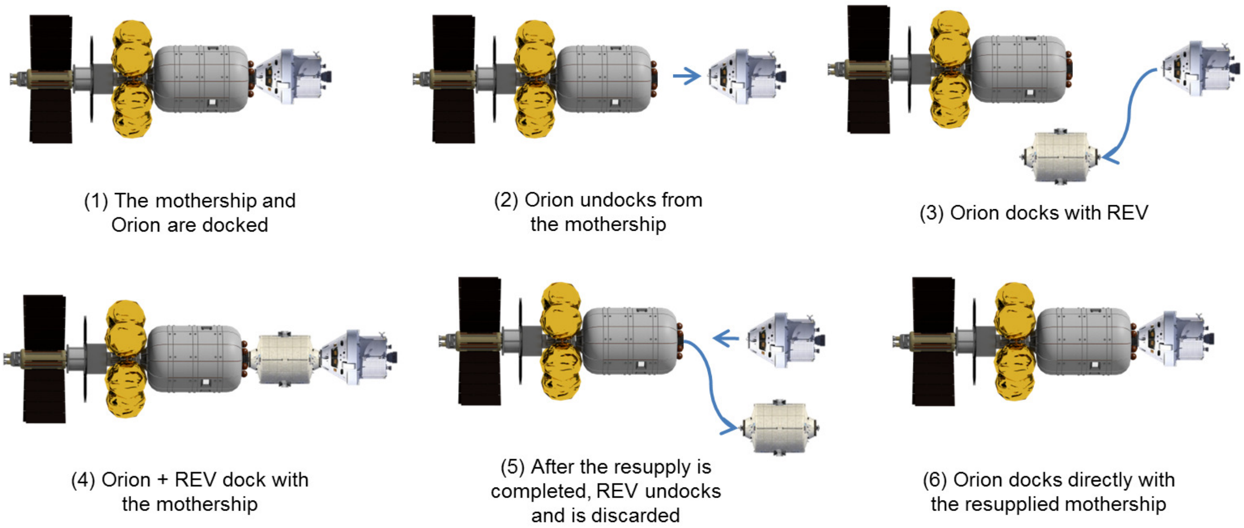


Figure 3. Main phases of the second two Resupply Expendable Vehicles, REV-3 and REV-4.

orbits is described in Appendix E. Additionally, MMRSD's interplanetary trajectory is shown in Figure 14 in Appendix F. Details regarding all of the major subsystems of IMaGInE are given in the following sections of the report. Table 1 summarizes the main phases of the entire mission.

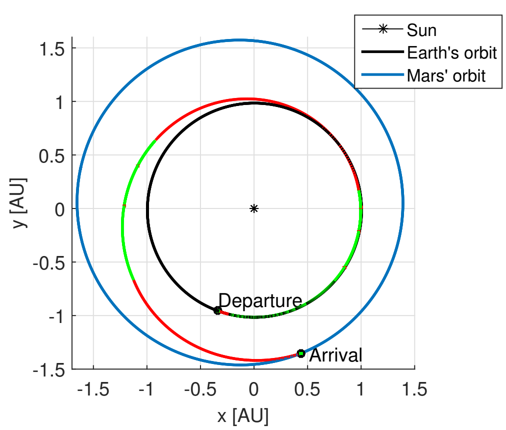


Figure 4. Earth-Mars. IMaGIne's interplanetary low-thrust outbound trajectory to Mars.

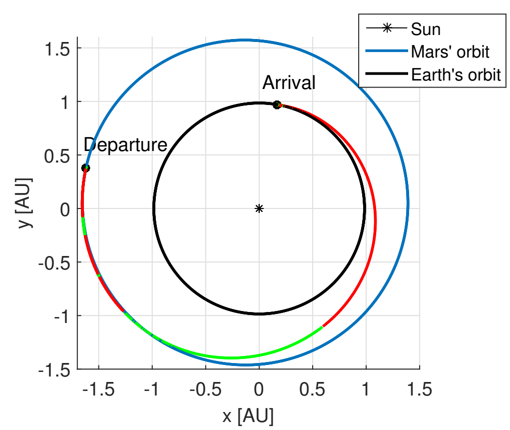


Figure 5. Mars-Earth. IMaGIne's interplanetary low-thrust inbound trajectory to Earth.

IV. Propulsion and Electrical Power System

To find an appropriate propulsion technology capable of bringing a spacecraft of more than 50 metric tons to a Martian moon and back (Δv > 12000 m/s), a trade-off was carried out for the three most promising and realistic technologies (see Table XIII in Appendix A). For this purpose, the two major characteristics of a propulsion technology, specific Impulse (I_{sp}) and thrust, have been divided into their resulting consequences for the mission architecture. I_{sp} is responsible for the payload fraction of a rocket and for the necessary IMLEO of an interplanetary spacecraft, while the thrust is mainly responsible for the time of flight of an interplanetary trajectory. By comparing these factors as well as TRL and safety of each technology, the most promising solution can be found. As a result of this trade-off, an electrically propelled spacecraft was found to be the best option.
In order to bring such a mass into LEO, a chemically propelled spacecraft would require either an infeasibly

Table 1. Mission analysis design parameters including margins. *U = Uncrewed; TC = Test Crew; MC = main Mission Crew.

Mission Phase	Initial Mass [t]	Final Mass [t]	Depart Date	fArrive Date	ToF [days]	ΔV [m/s]
LEO - Deimos (MMRSD)	84.80	56.60	28 Jan 2025	1 Apr 2030	1889	11413
LEO - Mars (MSPD)	54.40	13.60	18 Dec 2028	3 Sep 2029	259	3567
LEO - 60000 km (U)	99.91	89.90	6 Mar 2029	13 Nov 2029	252	5279
LEO - H2 (REV-3)	11.30	11.30	1 Aug 2029	1 Apr 2030	243	$\tilde{3}200$
LEO - 60000 km (TC+REV-1)	45.00	45.00	3 Nov 2029	13 Nov 2029	<10	4092
60000 km - H2 (TC)	138.04	133.95	13 Nov 2029	24 Feb 2030	103	1503
H2 - Earth (TC)	27.09	27.09	24 Feb 2030	6 Mar 2030	~10	390
LEO - H2 (MC+REV-2)	45.00	45.00	16 Mar 2030	26 Mar 2030	~10	4092
Stay at H2 (MC)	137.38	135.82	26 Mar 2030	15 Apr 2030	20	-
H2 - SOI Earth (MC)	135.82	133.93	15 Apr 2030	1 June 2030	47	700
SOI Earth - SOI Mars (MC)	133.93	124.43	1 Jun 2030	30 Aug 2031	455	3677
SOI Mars - Deimos (MC)	124.43	122.06	30 Aug 2031	29 Oct 2031	60	965
Stay at Deimos (MC)	136.34	109.53	29 Oct 2031	6 Oct 2032	343	-
Deimos - SOI Mars (MC)	109.53	107.44	6 Oct 2032	27 Nov 2032	53	965
SOI Mars - SOI Earth (MC)	107.44	99.23	27 Nov 2032	1 Jan 2034	400	3973
SOI Earth - H2 (U)	61.74	60.88	1 Jan 2034	23 Jan 2034	22	700
SOI Earth - Earth (MC)	27.09	27.09	23 Jan 2034	2 Feb 2034	~10	~400

high IMLEO, or an impractical number of launches. Even a spacecraft with a Nuclear Thermal Rocket (NTR) has a relatively low payload fraction compared to an electric propulsion system, thus resulting in a higher IMLEO. Nevertheless, it would be possible to design an interplanetary spacecraft using this technology. However, the TRL for an NTR is very low and the engines would exhaust radioactive material into the upper atmosphere. Moreover, there would be a massive radioactive contamination in the case of a launch failure. On the contrary, an electrically propelled spacecraft has the lowest IMLEO and gives the most feasible solution that can be launched into LEO. However, it has the lowest thrust and therefore the longest TOF, which has an unfavorable effect on the crew.

It can be seen that there is a trade-off between low IMLEO and low ToF. This suggests that chemical propulsion should be used for mission phases where the time of flight is most critical (i.e. crew transport), while electric propulsion should be used where IMLEO is most important (i.e. cargo transport). This leads to the concept of using electric propulsion for the mothership and using chemical propulsion to send the crew quickly and far. Since H2 can be reached by chemical propulsion in a quite short time and has an orbit with a high characteristic energy (C_3), it provides an appropriate place to dock the crewed spacecraft with the mothership. Thereby, the overall IMLEO can be reduced drastically while keeping the ToF for the crew at a reasonable length. This means that the crew will spend roughly one third of the whole mission time at Deimos. As a consequence, the concept that was implemented for IMaGInE was achieved by using both chemical and electrical technologies. This gives the outstanding possibility of keeping the IMLEO of a crewed interplanetary spacecraft in the range of the payload capability of a single SLS 1B and simultaneously reducing the mission duration for the astronauts by more than one year, compared to a solely electrical concept.

To implement this concept, four VASIMR engines are used to propel the mothership. These engines have one of the highest I_{sp} (5096 s) and thrust of all electric engines currently in development (5.76 N).[2] Due to the fact that each engine requires 200 kW of electrical power, a powerful Electrical Power System (EPS) is necessary. To find the most suitable technology for the EPS, a trade-off has been conducted. Table 10 in Appendix A shows that an EPS based on a nuclear technology is the best choice for the mission. This is mainly due to the very high weight specific power production and to the fact that the distance of the spacecraft to the Sun has no influence on power generation. For comparison, the solar constant decreases from Earth (1.367 kWm^{-2}) to Mars (0.5897 kWm^{-2}) by 57% and would therefore require solar panels with an area of almost 5 km^2 to support four VASIMR engines. Moreover, the technology of nuclear fission reactors is already flight tested and it enables a high expandability of the EPS. This is important because the required energy of an electrically propelled spacecraft is particularly sensitive to the spacecraft mass. Regarding safety, the chosen SAFE-400 nuclear fission reactor is passively safe in all launch or re-entry accidents and keeps subcritical even without any control. Moreover, it is not radioactive before operation.[3] Therefore, the propulsion and EPS concept used by IMaGInE is also much safer than an NTR, despite using nuclear technology.

V. Systems Engineering

All mass, power, and volume requirements, as well as costs, are assigned margins up to 20%, based on TRL and specifications. Finally, a system-wide margin of 20% is added. Design decisions are made in accordance to trade studies and well-defined subsystem requirements. The former are presented in Appendix A, while the latter can be traced to Top-Level (TL) requirements and competition Ground Rules (GR), which are given in Table 15 and 16 in Appendix B. This allows for a complete assessment of the overall infrastructure, ensures fulfillment of the mission, and avoids over-design.[5] Based on derived requirements and NASA standards,[6] a risk analysis has been performed to ensure failure modes have been mitigated (see Appendix C). The test schedule and development plans have been established based on TRL, launch manifest, and synergies with existing programs. The critical technologies, their estimated initial and targeted TRL, and the implementation of the development program are shown in Table 2. None of the used sources are older than 12 months to ensure all information is current.

Table 2. Development of critical technologies.

Technology	TRL	Implementation
ECLSS - Torpor	3 - 8	Currently under development with NASA support.[7] Use in study similar to Mars 500 for testing (could involve ISS).
Science - Space Solar Power	4 - 8	Currently under development by Caltech and Northrop Grumman Corporation.[8] Tests can be performed in LEO or with regard to planned moon missions.
Science - MAN Weather Station	6 - 9	Modified version of existing weather balloons.
Science - Moon Hoppers	5 - 7	Can be tested during ARM and Earth's Moon robotic missions.
Deimos Science and ISRU	5 - 7	Can be tested during ARM.
Mars Science and ISRU	5 - 7	Can be tested during ARM.
Propulsion - VASIMR	3 - 8	Currently under development with NASA support,[9,10] with goal of testing the engine on the ISS.
EPS - Safe-400 Fission Reactor	3 - 7	Basic technology exists. Most efforts have to be expended to increase reliability and safety.

The development schedule is shown in Figure 6. As human factors are of paramount importance, and a proposed, novel technology is expected to affect the crew, an extensive test environment is suggested, similar to the Mars500 experiment.[11] This environment should be created to show the feasibility of a continuously crewed mission lasting 3.6 years, test the continuous operation of the torpor units, test the torpor crew rotation cycles, study the effects on the astronauts, and determine the demand of maintenance required by the torpor units. Additionally, the mental stability of the conscious astronauts can be evaluated as well as the operational skills of the crew regarding the spacecraft after such a long time. The test environment runs from 2021 to 2025. Thus, there would be 4 years during which to implement knew knowledge and make adjustments to the actual mission before the test crew launches.

The IMaGInE Mission will launch an overall total of 295.6 metric tons to conduct the proposed mission, using two Falcon Heavys and four SLS Block 1Bs. The science mission requires 10.4 t, which gives a margin of 27% on the launch capacity. The crewed mission requires an overall 287.4 t, which gives a margin of 10%

on the launch capacity. Thus, the mass requirements are satisfied by the available launch capacity and Δv. The volume requirements have been considered in the habitat and service-module design, and the power requirements are met by the SAFE-400 reactors and the Space Solar Power satellite. Budget summaries are given in Table 3 and 4.

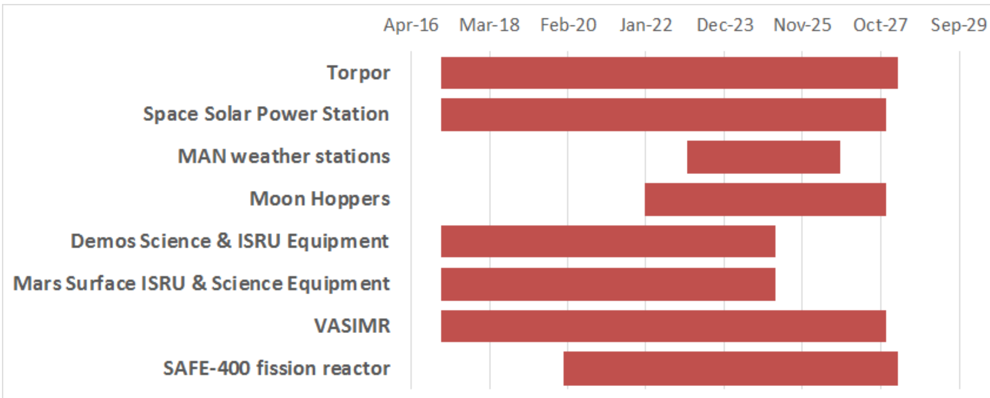


Figure 6. Schedule for the development program of critical technologies

Table 3. Science Budget

	Mass [t]	Volume [m^3]	Power [kW]
Total	10.4	22.8	289.8
Total + 20%	12.4	27.4	347.8
Provided	13.6	116	350

Table 4. Crewed Mission Budget

	Mass [t]	Volume [m^3]	Power [kW]
HERMES + HARMONIA	155.7	149.7	482.5
Orion	2 x 25.8	-	-
Resupply	86	-	-
Total	293.3	149.7	482.5
Total + 20%	351.96	179.6	579
Provided	315.5	349.5	600

VI. Attitude and Orbit Control System and Landing/Ascent at Deimos

The main objective of the Attitude and Orbit Control System (AOCS) is to provide spacecraft navigation and orientation maneuver capabilities to point the spacecraft at desired targets based on mission requirements. It is designed to minimize fuel consumption following the guidelines of the innovative risk-informed design process of NASA that allows the team to design a vehicle with the best safety and reliability.[12]
Propulsive maneuvers, crew activities, fuel slosh, and thruster misalignment are some disturbances that must be corrected to keep the desired attitude within an accuracy of $<0.1°$ in each axis. This section presents a preliminary design of AOCS that complies to the requirements and constraints of the IMaGInE Mission and NASA-ESA standards. The mothership and Orion (with its service module) are both three-axis stabilized and are provided with a Failure Detection Isolation and Recovery (FDIR) system. Different AOCS modes of performance have been selected mainly depending on the mission phases and pointing requirements. The

error feedback is used in every AOCS mode since it provides the desired amount of performance and robustness against parametric and model uncertainties.

In order to determine the attitude of the spacecraft, different Commercially-off-the-Shelf (COTS) sensors have been selected. Two sets of three Sun sensors (cold redundancy) by Honeywell have been selected. In terms of FDIR, the three Sun sensors are simultaneously on (hot redundancy). This ensures correct attitude determination should one unit fail. Primary and backup Inertial Measurement Units (IMUs) (Honeywell HG1900) measure changes to the spacecraft attitude as well as any non-gravitationally induced changes to its linear velocity. Each IMU is a combination of three accelerometers and three ring-laser gyroscopes. Two autonomous star trackers manufactured by Ball Aerospace are co-aligned at 90° to provide 3 axis inertial attitude measurements, each with a field of view of 8 by 8 degrees, used in cold redundancy.

Trajectory Correction Maneuvers (TCMs) are performed mainly during orbital maneuvers for station-keeping purposes and momentum unloading. The actuators selected for this purpose are two sets of 4 Control Momentum Gyros (CMGs) and 32 Reaction Control System (RCS) thrusters capable to perform TCMs and fine attitude and orbit control maneuvers. A trade-off study among different types of thrusters to compare the performance of innovative and classical thruster technologies can be found in Table 12 in Appendix A. A pressure-fed integrated RCS using LOX and methane (LCH_4) thrusters has been selected. Aerojet 100-lbf thrust LOX/LCH_4 was selected due to its high I_{sp} qualities (317 s), non toxicity, long term storability, suitability for ISRU and the possibility to use the crew's biowaste products.[13]

Landing and Ascent at Deimos

The mothership + Orion will land on the surface of Deimos with a primary goal of landing precisely and safely. It will rest on a four-legged landing gear placed on Orion's service module (Figure 7). The spacecraft will include an innovative, autonomous navigation system that will be capable of landing without crew assistance and recognizing and avoiding hazards such as craters and boulders; this system includes three Light Detecting And Ranging, or LiDAR, sensors and navigation cameras.[14] The mothership will perform a soft-landing, and assuming uncertainties, only low impact velocities will occur at touch-down.

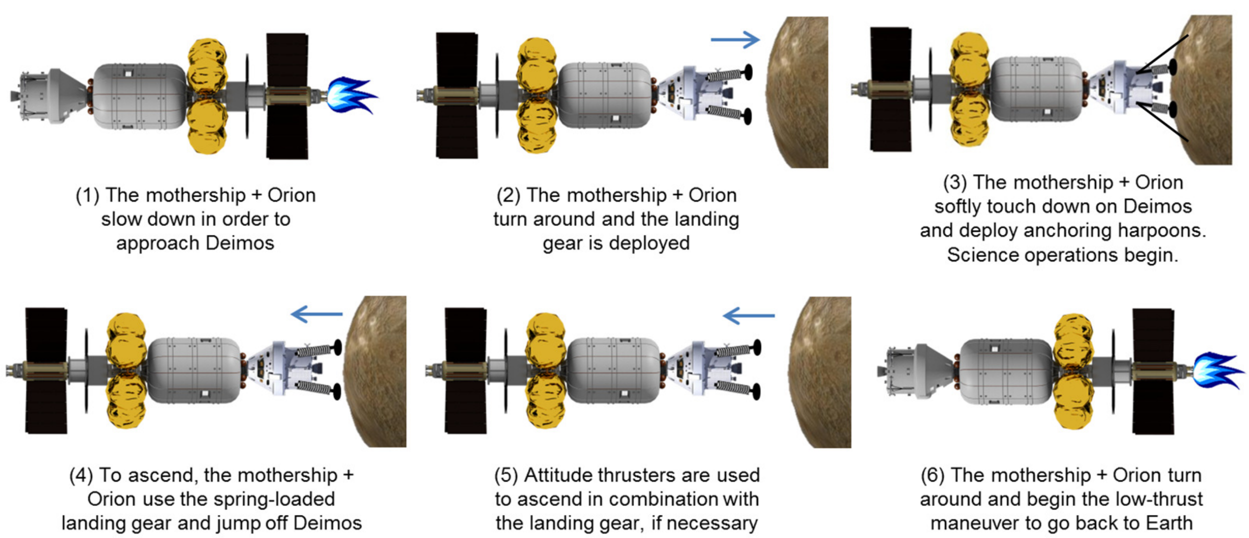


Figure 7. Main phases of the landing (1-3) and ascent (4-6) at Deimos.

While approaching Deimos, a Δv will be applied by HERMES to induce a near vertical descent the surface. The vertical thrusters will be turned off at an altitude of approximately 100 m. From this point, just small thrust corrections will be performed down to an altitude of 10-20 m, at which time it will have near-zero velocity. In order to prevent the thruster exhaust from contaminating Deimos regolith, the spacecraft will free fall from this point.

Due to Deimos' low gravity, re-bouncing becomes a significant issue and anchoring is required.[15] Thus, the four landing legs will include ice-screws and an innovative damping system with the capability not only to smooth the impact, but also to store potential energy that can be used at the initial phase of the ascent. This is to prevent the use of RCS thrusters that could contaminate the moon's surface. Therefore, four anchoring ropes with harpoons will be fired to help keeping the local vertical. RCS is left as a backup solution in case the energy stored in the landing legs is not enough to reach escape velocity. HERMES' propulsion system has not been considered for ascent since the RCS thrusters give enough thrust for the ascent from Deimos. A trade-off concerning landing strategies is summarized in Table 11 in Appendix A.

VII. Environmental Control and Life Support System and Human Factors

During the journeys to and from Deimos, crew members will make use of torpor. Torpor, which uses therapeutic hypothermia, allows the crew to enter an unconscious state of decreased body temperature and metabolic rate. Placing humans in this state reduces the consumption of life support resources, production of waste, and will avoid many of the psychological concerns associated with long-term spaceflight.[16] This reduction in consumables allows for significant mass savings. On average, a crew of four can save about 55 kg of consumables per day using torpor. Figure 8 shows the minimum, maximum, and average savings of consumables per day using torpor.

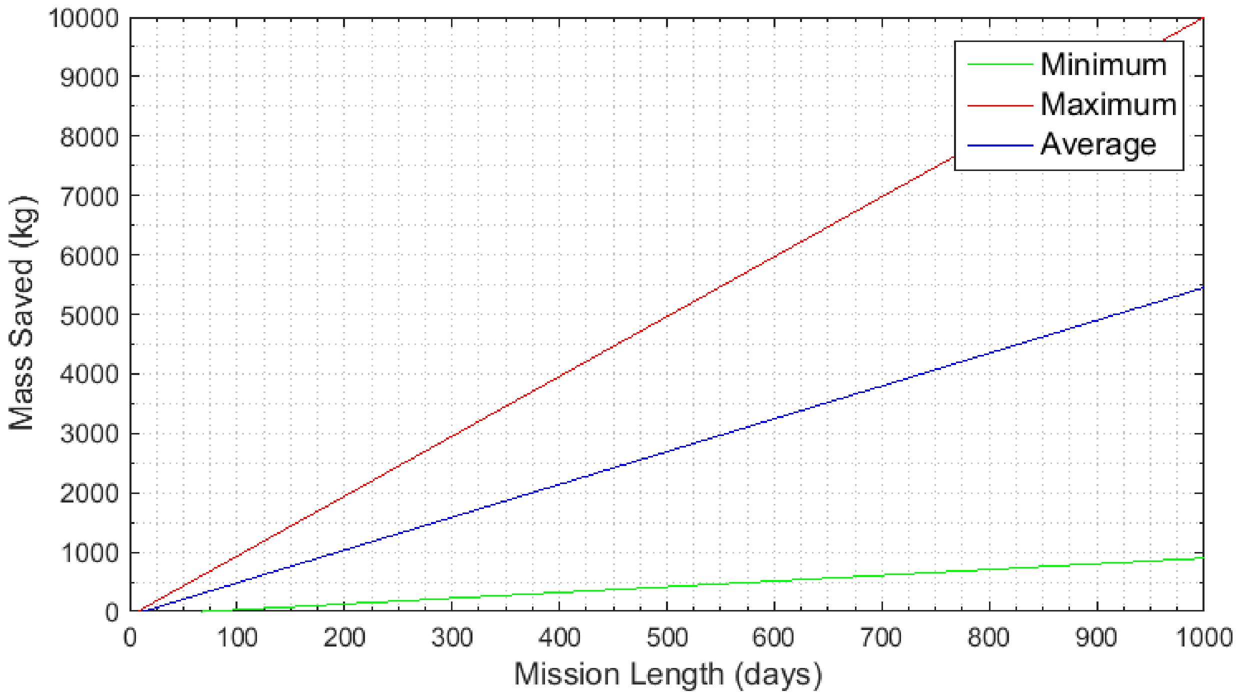


Figure 8. Torpor mass savings per day over mission duration.

During the course of the mission, astronauts will be placed in a rotating torpor state; all crewmembers will be awake for 4 days at a time followed by 5-11 days in a torpor state (including induction and awakening from torpor). During the trip to and from Deimos, one crew member will always be awake to manage communications with the ground, administer regular system checks, monitor crewmembers' vital signs, and aid in the torpor-induction and awakening of other crewmembers. In Figure 9, an example of the torpor schedule can be seen. Staggering torpor schedules as seen will allow for each crewmember to constantly be in the company of different crewmembers during their times awake. This will improve psychological states for each crewmember. Allowing each crewmember to be alone for part of a day during their active state will also prevent the stresses associated with constant companionship during the long mission to Deimos.
Risks associated with normal microgravity spaceflight including bone density loss and muscle atrophy can be mitigated through the use of pharmaceuticals and physical training in workout facilities on board HARMONIA. The risks and their associated mitigation techniques for the use of torpor are given in Table 18 in Appendix C. The long mission to Deimos will require one crewmember to be a flight doctor. This crewmember will be able to track other crewmembers health during the mission. This will mitigate risks associated with torpor and ensure any sickness or injury can be taken care of on-board the spacecraft.
Human patients that have undergone multiple cycles of therapeutic hypothermia showed no negative effects

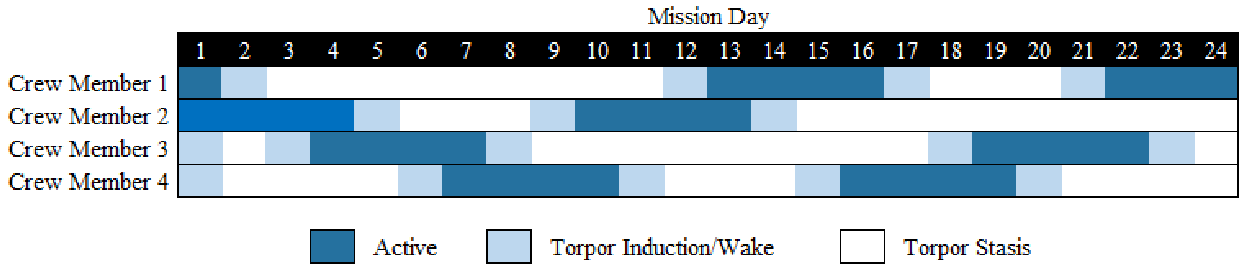


Figure 9. Torpor rotating schedule example

from the cyclic procedure in short-term or long-term timeframes.[16] Spaceworks Engineering, Inc., the company who completed the initial evaluation of torpor habitats for astronauts during long-term spaceflight, have recently been awarded $500,000 from NASA to further their research and complete a Phase 2 study. This research will aid in the advancement and readiness of this technology.

In order to further identify and reduce the risks associated with torpor, testing can be completed prior to the mission both on Earth and on the ISS. Patients can be placed into torpor states in bed-rest studies in order to simulate the effects of micro-gravity and torpor on the body while being under constant observation on the ground. These tests will help identify and reduce any further risks not known. Isolation studies can also be completed with torpor. Four patients can be placed into isolation with one another while being placed in a torpor cycle. Isolation tests will help identify the benefits and psychological effects of rotating torpor cycles in an isolated environment. A torpor module can also be placed in an inflatable module on board the ISS to fully test the effects of multiple day torpor cycles in succession in a microgravity environment. All of these tests will further the readiness of the torpor technology and mitigate the risks associated with it.

Orion is equipped with a CO_2 and Moisture Removal Amine Swing-bed (CAMRAS) atmospheric revitalization system. Orion is also equipped with an active thermal control flow system and trace contaminant system. A water recovery system will need to be integrated into the Orion capsule for the long-duration travel to and from Deimos. HARMONIA, modeled after Bigelow's BA-330 habitat, will accommodate the torpor pods for the crew. This inflatable environment will be equipped with the Sabatier carbon dioxide removal system, JPL E-Nose for fire detection, fine water mist fire extinguishers for fire suppression, a Vapor Phase Catalytic Ammonia Removal (VPCAR) system for water purification and recycling, and an Oxygen Generation Assembly (OGA) that is currently on the ISS. The trade study completed to determine the optimal CO_2 removal system can be found in Table 14 in Appendix A.

For launch, re-entry, and landing on both Earth and Deimos, crewmembers will use Modified Advanced Crew Escape Suits (MACES). The MACES suit provides a pressurized environment for the crew in the event of an emergency depressurization of the Orion capsule. This will allow the crew to initiate a launch-abort scenario during launch, or give enough time for the crew to move to HARMONIA if away from Earth. The MACES suit also functions as an emergency Extra-Vehicular Activity (EVA) suit. During EVA operations, the Z-series space suit will be used due to its advanced life support and mobility capabilities. The Z-series space suit will allow crew to complete all required work on the surface of Deimos. Additionally, the margins on consumables allow astronauts to perform emergency EVAs to perform spacecraft repairs while maintaining the nominal mission profile, despite having to depressurize and re-pressurize Orion.

VIII. Communications

The communications system consists of two parabolic, high-gain antennas each with a diameter of 3 m. In addition, four omni-directional antennas are installed to ensure constant telemetry, tracking, and command. These antennas are designed to work with X-band, the current standard of the Deep Space Network (DSN) and ESTRACK for interplanetary missions.[17] Moreover, the spacecraft will be equipped with a UHF communication system for teleoperation activities on Deimos and Mars and to allow for relay connections with nearby probes. This also enhances safety through redundancy and would allow for more data to be sent to Earth. Figures 10 and 11 show that the downlink rate to Earth using the RF link is low during the astronauts stay at Deimos. Using the 34 m antennas, available in both the DNS and ESTRACK network, the downlink can drop to as low as 24 kbit/s, assuming 100W transmitter power. This could be enhanced by using stronger transmitters such as the DSN 70 m antennas, K-band, or optical communications. The latter two are currently under development with promising results.[18] Nonetheless, assuming the DNS network can be used at least as much as MRO is using it now,[19] an average of 25 images per week, plus an estimated 1 kbit/s for astronaut monitoring, 1 kbit/s for TTC, and 14 kbit/s for general communication can be allocated using QPSK modulation. These values may be increased as needed.

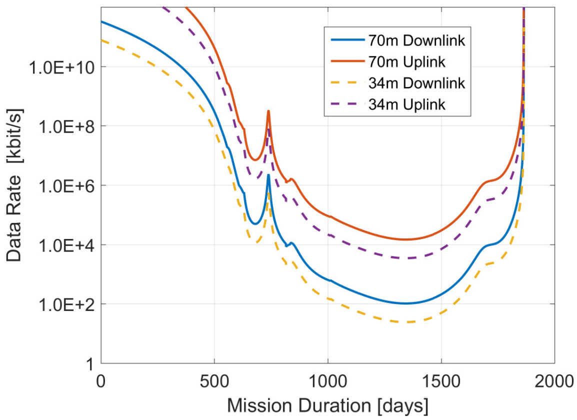


Figure 10. Data rate over the entire mission duration using X-Band and a 100W spacecraft transmitter.

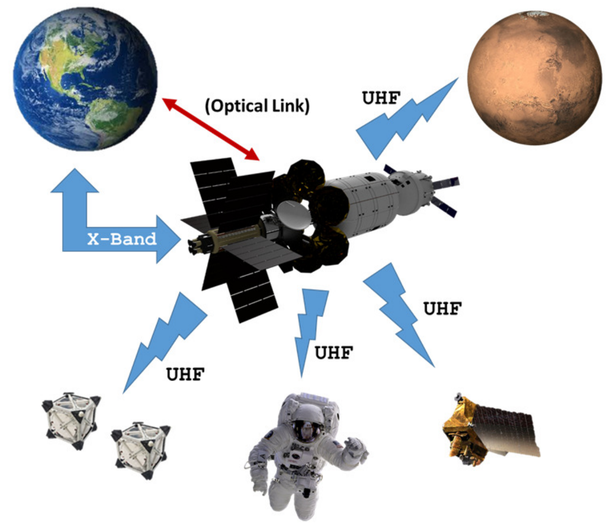


Figure 11. Communication pathways during the mission. An optical link would be desirable to increase data rates significantly.

IX. Science and Robotics

The primary science and technology goals of the mission are to enable future crewed missions to the surface of Mars with interest in colonization. To achieve this, the mission deploys a network of science stations, demonstrates feasibility of fuel, water production, and 3D printing of large structures on the surface of Mars and its moons. Power will be provided to all ground assets from Space Solar Power (SSP) stations. Further science will be conducted by Moon Hoppers at the surface of Phobos, and by astronauts on Deimos. Human exploration is included in the mission to provide a subjective perspective of the inhabitability of the Martian system, ensure the most interesting aspects of the celestial bodies are being observed, and provide quality control in data collection. Pre-existing assets on the Martian ground that are still in working order, such as ExoMars, will be teleoperated from Deimos for technology demonstration. The mass of scientific payload is summarized in Table 5. To interact with robotics deployed at Deimos, the crew will utilize Shape Memory Alloy (SMA) beams. These are lightweight structures than can be easily extended or stored due to their thermal properties.[20]

Martian surface Analysis Network (MAN)

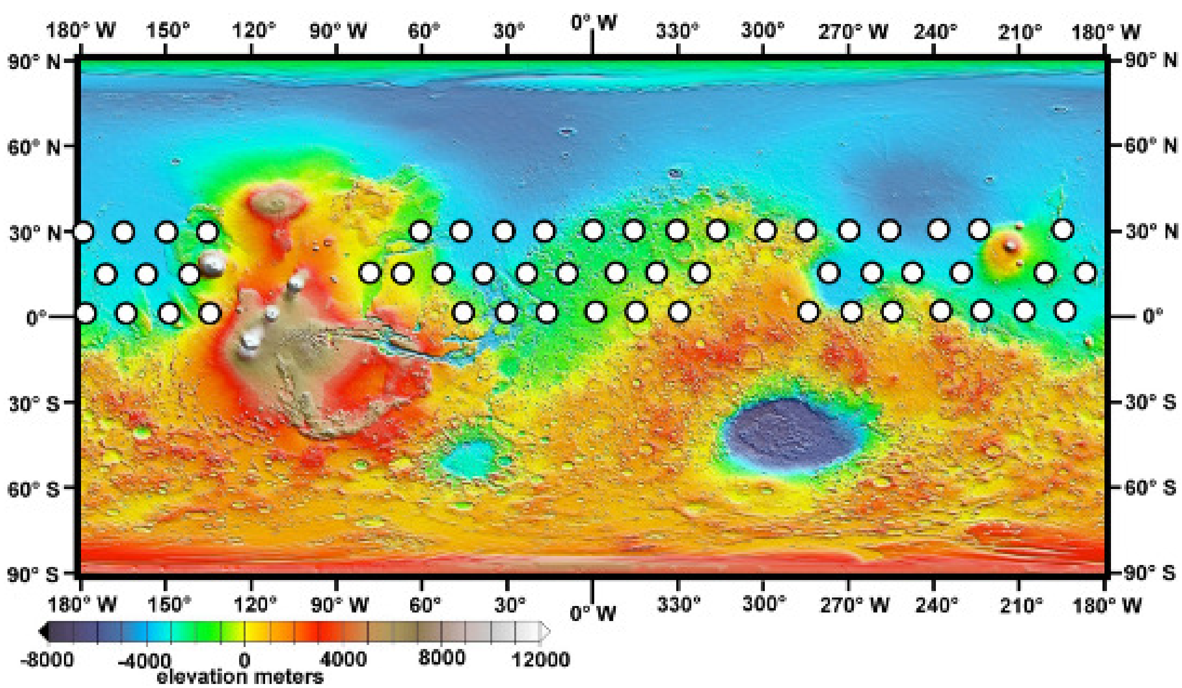
Three evenly spaced latitudinal profiles of 54 science stations will be landed between 0° and 30°N (Figure 12). Their locations will cover most of the area that meets landing requirements (both latitude and elevation) for future human missions. Each lightweight station (36.5 kg) is released in low Mars orbit and landed via airbags and retrorockets. One purpose of this network is to characterize Martian surface weather and soil properties at an unprecedented spatial and temporal resolution, to help identify optimal landing sites and enable the human exploration of Mars. Each station includes a seismometer, ground heat probe, temperature, wind (velocity+direction), and humidity sensors, a 360-degree panoramic camera, radiation sensor, a microscopic imager to determine regolith grain size, and a soil and organics test instrument to assess the nutrient and organics content of local regolith. Finally, each station will have a data transmission antenna and a microwave receiver for receiving power from orbit. Entry, Descent, and Landing (EDL) and

Table 5. **Mass summary for scientific equipment.**

Equipment	Mass [kg]	Number	Total Mass [kg]
Space Solar Power Station	370	3	1110
MAN stations	71.5	54	3860
Moon Hoppers	60	2	120
Moon Hoppers Propulsion Module	244	-	244
Deimos Science and ISRU Equipment	800	-	800
Mars Surface ISRU and Science Equipment	2400	-	2400
Sky Crane for Mars Surface Equipment	750	-	750
Total	-	-	$\sim$ 9284

structural mass is based on the Beagle 2 lander mass budget,[21] which yields a revised total mass of 71.5 kg/station. A detailed mass breakdown can be found in Appendix D. To reduce cost and development time, the MAN stations use many heritage components. The cameras are inherited from the Mastcam on MSL and the heat probe from INSIGHT. The organics detector is reused from the Sample Anaysis at Mars (SAM) instrument suite on MSL.

Figure 12. Example grid for the Martian surface Analysis Network, designed to characterize possible future landing sites for a manned mission to Mars.



Landers were favored over orbiters because the latter are unable to directly measure many of the ground surface properties the mission seeks to characterize, such as radiation levels, geothermal gradients, nutrients, perchlorate, volatiles, and dust contents of the soil. For the same mass, landers also provide data from 54 locations, as opposed to less than half a dozen if Curiosity-like rovers are used. Cameras will allow imaging of assets of the ground (e.g. rock sizes/thermal inertia, relevant to building/shielding) that are below HiRISE resolution[a]. The MAN is critical for identifying optimal landing sites, allowing full coverage of the

[a]High Resolution Imaging Science Experiment onboard MRO. HiRISE offers the highest resolution of the Martian surface to this date, with a pixel size of about 30 cm at best, and has a relatively small footprint due to its high resolution.

latitudinal region suitable for landing, and thus, paving the way to human exploration. In contrast, landing three isolated rovers requires preselecting landing sites from a fraction of the assets that are measurable from orbit, and limits the range of future opportunities.

Table 6. MAN Station Mass Estimate.

Subsystem	Mass [kg]
Probe	35
Lander	24
Science Payload	12.5
Total	~ 71.5

Moon Hoppers

Low gravity results in low friction, making it impossible for traditional rovers to drive safely on these celestial bodies. Thus, the Highland Terrain Hoppers (Hopter), jumping robots driven by three independent actuators consisting of electric motors, gears, and springs will be used. These robots have a reversible main body and three firing legs that allow them to hop and avoid obstacles much larger than their own size. Moon hoppers are designed to recover from falls and impacts, which are common with this method of maneuvering.[22–24] When utilizing moon hoppers, science equipment will be designed and mounted in a way that protects it from harsh conditions. Two moon hoppers will be deployed on Phobos to characterize its composition and structure, with one characterizing spectroscopically blue terrain and the other characterizing spectroscopically red terrain.[25] In addition to ISRU capabilities, their payloads include an alpha particle X-ray spectrometer for chemistry, X-ray diffraction spectrometer for mineralogy, microscopic imager, spectral camera, and a georadar. The total mass of each moon hopper is 60 kg.

Space Solar Power (SSP) Stations

Three SSP stations (370 kg each) capable of generating 200 kW each will orbit Mars providing continuous power coverage to all assets on the ground. In development at Caltech, these ultralight structures[26] allow solar energy to be concentrated onto thin photo voltaic (PV) panels, then beamed down to the Martian surface as microwaves using a phased array antenna. Using foldable booms, each can be packaged into a 1.5 m high and 1 m diameter cylinder. The phased antenna approach ensures power is generated and converted to microwaves locally, rather than incurring transmission losses. Current calculations show specific input power up to 6.3 kW/kg in Mars orbit. Including losses, 200 kW/station is eminently feasible. These stations will also provide power for future missions, eliminating the need for nuclear reactors. They will also act as relays, sending data back to Earth.

In Situ Resource Utilization (ISRU)

A miniaturized JPL ATHLETE robot[27] (450 kg) consisting of two fully independent three-limbed robots (Tri-ATHLETEs) will be used to move ISRU equipment around at a primary landing site on the Martian surface. The primary landing site will have autonomous fuel production units. These will take 50 kg of H_2 feedstock and turn it into one metric ton of CH_4 and O_2. In addition to fuel production, the mission will bring 60 kg of raw materials and construction equipment such as scoops, levelers, and a large 3D printer. These materials and tools will allow for the assembly of large structures that will demonstrate the technology needed for habitats, the building of a storage dome to protect equipment from dust storms, and the 3D printer will aid in equipment construction, repair, and replacement.

On Mars, the miniaturized ATHLETE will be able to carry up to 400 kg in payload. While not carrying any payload, the robots could be used to scout the area. Since the time delay is much smaller than while operating from Earth, it can enable some new activities never before done with rovers. On Deimos, the astronauts will study the moon's geology and look for hydrated minerals. If found, these minerals will be crushed, baked and then liquid water extracted via a centrifuge. The water will be split into H_2 and O_2 and tested for its potential use in rocket fuel. The feasibility of utilizing processed regolith as heat shields for Martian landings will also be investigated.

Teleoperation

Astronauts on Deimos will be able to teleoperate rovers on Mars because of the lack of a time delay. Teleoperation will enable Martian rovers that are still operable, such as the ExoMars rover, to be reused. This will allow for nearly real-time exploration of Mars and the examination of human-robotic interaction. Though existing rovers are slow, the lack of a time delay will make the operating process much faster.

Traditional Mars rovers are designed to move slowly due to time delays, but since this mission aims to send humans near Mars, the new, Tri-ATHLETE robots will be designed to move much faster, enabling astronauts to explore more of the Mars surface than ever before.

X. Thermal Control System

The main purpose of the Thermal Control System (TCS) is to cool the four SAFE-400 nuclear reactors which produce a thermal power of 3.84 MW. The core temperature of each reactor is ∼1200 K and it is assumed that the incoming coolant temperature shall not exceed a temperature of ∼500 K. This results in a maximum radiator temperature of ∼700 K. On this basis, the effective radiator area can be calculated to an area of 288 m^2. Assuming a standard radiator geometry of 6 radiator panels, this results in 4 m by 6 m radiators. This gives a reasonable radiation geometry and mass estimates for such a large amount of power. This is possible due to the fact that a relatively high radiator temperature is used.

XI. Radiation Shielding

On the surface of the Earth humans are shielded by Earth's magnetic field and are only exposed to non-ionizing radiation like UV rays. On the other hand, in space there are ionizing radiation and solar energetic particles. The former can have a high level of energy while the latter are released by the Sun and have a lower energy. Various types of radiation can cause radiation sickness and other acute and chronic effects. The acute effects can be nausea, vomiting, and fatigue. The chronic effects are the result of a longtime exposure to radiation that can manifest themselves even decades after the exposure (e.g. cancer).

In order to protect the crew from harmful radiation, spacecraft structures must be strengthened. Thicker walls and solid shields are the best way of protection, but are also the most massive solutions. Spacecraft walls made of heavy and rigid materials would make the overall mission unfeasible if a perfect shielding is desired. Therefore, the mass which is already present on board will be used to shield the astronauts. For example, cabin material can be moved to build a temporary shelter in case of high radiation events. These materials include all movable parts of the spacecraft such as supplies, equipment, launch and re-entry seats, water, and food. Water is especially a good material that can shield astronauts from radiation.[28] Thus, no additional mass is added to the system. HARMONIA is featured with an approximately 0.46m thick hull that provides shielding against radiation and also against ballistic particles.[29] The combination HARMONIA and Orion provides an acceptable shielding concept.

Another radiation source is the SAFE-400 housed in HERMES. Since materials with a high concentration of hydrogen provide the best shielding against radiation,[28] water tanks and VASIMR's propellant tanks are placed between the crew and the nuclear reactors in order to utilize them as shielding.

XII. Cost

Initial cost estimates are based on mass, heritage, and the NASA AMCM including a 2% inflation rate. The operations costs are estimated from the ISS program.[30] The total cost given in Table 7 is for the entire 20-year program, including development and a total of B$10.4 FY2016 in operations cost over eight years.

After an inflation adjusted analysis of NASA's budget in accordance with the given ground rules (see Appendix B), this mission should have access to a total of approximately B$102 FY2016, with more than B$9 FY2016 per year starting in 2026. Currently the mission would use 31% of the total budget, thus there is a large margin to absorb additional costs. Development costs are estimated using guidelines provided by,[31] the NASA Advanced Mission Cost Model (AMCM), and heritage. Additionally, information available from press releases with regard to existing programs were considered for comparison and baselining. The resulting amounts are shown in Table 8, including a 65% margin for wrap costs. A short reasoning and information on which sources were used are also provided. The total yearly mission cost is shown in Figure 13.

XIII. Conclusions

The mission design presented in this paper was created with the objective of being a sustainable and evolvable mission that makes use of a series of innovative technologies. In fact, the mothership was designed

Table 7. **Cost Budget.**

	Cost [M$]
Phase A Wrap Cost	28
Phase B Wrap Cost	331
Phase C/D Wrap Cost	2,253
Development Cost + 20%	5,712
Spacecraft + 20%	3,755
Launcher Cost + 20%	2,590
Ground Control & Operations 8 years	10,400
Total	24,974
Total Inflation Corrected (FY2016)	31,734

Table 8. Development of critical technologies. M$ in FY2016.

Technology	Cost [M$]	Source of Estimate
ECLSS - Torpor	234	Based on NASA funding + 1 launch + 3.6-year test environment
Science - Space Solar Power	876	Northrop Grumman funding budget and AMCM + 1 launch
Science - MAN Station	25	Development cost equal to three units built
Science - Moon Hopper	57	Development cost equal to building demonstrator
Deimos Science and ISRU	405	AMCM + 1 launch
Mars Science and ISRU	394	AMCM + 1 launch
Propulsion - VASIMR	1,910	NASA funding budget and AMCM + 1 launch
EPS - Safe-400 Fission Reactor	855	AMCM + launch

with the intent of being a reusable spacecraft for exploring the moons of Mars, and allow humans to eventually arrive in low Martian orbit and then descend onto the red planets surface. The mothership is nominally kept in a parking orbit near EML2, which favors the use of the spacecraft for missions taking place both in cis-lunar and deep space. Resupplies can be performed to replenish the spacecrafts consumables and propellant for future missions in a similar way to how REVs are used. Another innovative trait of the mission presented in this paper is that the use of hybrid propulsion (chemical and electrical), combined with the trajectory optimization technique described in Appendix E, allows the IMaGInE mission to take place with the use of a relatively lightweight spacecraft.

This mission is aimed at enabling future exploration of Mars. In fact, the assets delivered to the Martian moons, such as the moon hoppers, and onto the Martian surface, such as the MAN stations and 3D printing equipment, are designed with the idea of being used for future missions in the Martian system, not simply for a one-time use during the IMaGInE mission. Future missions will thus further our knowledge of Mars, Phobos, and Deimos and they will favor the establishment of human colonies on the red planet.

Acknowledgments

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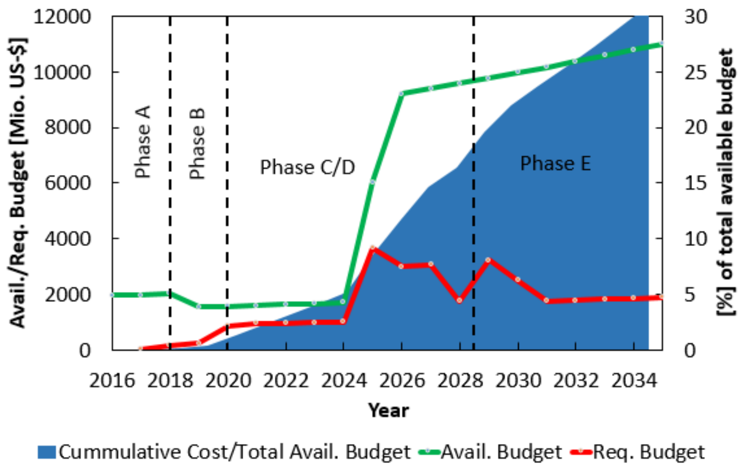

Figure 13. The total mission cost (FY2016) per year is shown together with the available budget and the cumulative cost divided by the total available budget. Approximate time frames for the different mission phases are separated by vertical lines. Currently 31% of the total available budget from 2016-2035 is required.

in September 2016. The team would also like to thank the Penn State committee of professors and experts that helped the team through a Preliminary Design Review (PDR) prior to the 2016 RASC-AL Forum.

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Appendix A: Trade-off Matrices

This appendix provides the trade-off matrices that are the result and justifications of various trade studies for subsystems and general mission decisions.

	Phobos			**Deimos**		
	Rationale	*Pro*	*Con*	*Rationale*	*Pro*	*Con*
	Double the gravity, easier for surface operations and ISRU	3		Very subdued surface, likely mantled in regolith, not much access to bedrock		3
	Thich regolith (200 m), might be harder to get to bedrock		2	Hemispheric-size crater, may provide access to the subsurface	5	
	Might be plastered with Mars material	2		Less probability of finding Mars material		3
	More likely to be differentiated	3		Less likely to be differentiated		3
	Large impacts (Stickney crater) and pits provide access to the subsurface	5		From Viking encounter seems to be smooth at 1m scale, i.e. less risky to land on a large rock	3	
	Less frequent line of communication to Earth		2	More frequent direct line of communication to Earth because, as viewed from Deimos, Mars does not occult Earth as frequently	2	
	Orbital period is 8 hours, more direct line of sight to Mars	5		Orbital period is 30 hours, limiting the amount of visibility with the Martian surface per sol		4
	Needs a ΔV of 1570 m/s more than to get only to Deimos (same amount of the final Trans-Mars-Injection)		5	No need for additional ΔV of 1570 m/s	5	
	From Phobos assets can be teleoperated on Mars up to 64.8 deg latitude		3	From Deimos assets can be teleoperated on Mars up to 80.2 deg latitude	3	
	Short communication passes to sites on Mars (4 hours)		3	Longer communication passes to sited on Mars (2.5 days)	3	
	Radiation: Mars fills 3.4 % of the 4 π steradian sky	2		Radiation: Mars fills 0.5 % of the 4 π steradian sky		2
	Worse illumination conditions than Deimos		3	Better illumination conditions than Phobos	3	
TOT		20	18		24	15
Pro/Con		**1.11**			**1.6**	

Table 9. Trade-off for Mars moon. 1- not important, 5 - important.

Propulsion technology	**resulting payload fraction***	**IMLEO mass***	**resulting Time of Flight****	**TRL**	**Safety**	**Final ranking**
Chemical	- -	- -	+	++	+ +	2.
Nuclear thermal	-	-	+	- -	- -	3.
Electrical	++	++	-	+	+ +	1.

EPS technology	Max power generation	Influence of Sun distance	Weight specific power	TRL	Expand-ability	Safety	Final ranking
Solar	+	-	+	++	0	++	2.
Nuclear	++	++	++	0	++	0	1.
Stored	-	++	- -	- -	- -	-	3.

Table 10. Trade-off for EPS technologies

Type of landing	Bounce risk	Damping energy stored	Fuel consumption	Contamination	Final ranking
Soft	- -	+	- -	-	1.
Hard	- - -	++	-	- - -	2.

Table 11. Trade-off soft vs. hard landing on Deimos

Type of Thruster	Performance	Toxicity	Storing	Refueling	Final ranking
Hydrazine	+++	- -	+	-	2.
Green Biowaste (Oxygen/methane)	++	++	++	++	1.

Table 12. AOCS thruster selection

	Mass	Redundancy	Soil/Radiation Measurement	Surface Area Covered	Final ranking
Single Orbiter	++	-	- -	++	2.
Rovers x3	+	-	++	- -	3.
Landers x54	+	++	++	++	1.

Table 13. Trade-off on type of science surface assets

	Sabatier	**Bosch**	**LiOH**
Inputs	CO_2, H_2, [$H_2/CO_2 = 4.5$], Heat	CO_2, H_2, heat	H_2O, CO_2, N_2, O_2, LiOH
Outputs	CH_4, heat, H_2O	C, H_2O, heat	H_2O, N_2, O_2, CO_2, H_2O
Efficiency	96%	N/A	N/A
TRL	6	4	8
Operability	Autonomous. Only maintenance required involves part replacements after long durations of mechanical wear.	Integration more complex than Sabatier. Catalyst cartridge must be periodically replaced by crew members.	Non-regenerable. The reaction that occurs from the LiOH sorbent is irreversible. The crew will need to replace LiOH cartridges daily making this a poor interface for the crew.

Table 14. CO_2 Removal Trade Study.

Appendix B: Ground Rules and Top Level Requirements

Mission Statement:
The IMaGInE Mission (Innovative Mars Global International Exploration Mission) will deliver a crew of four astronauts to the surface of Deimos for 300 days during the years 2028 and 2034. The crew will perform surface excursions, technology demonstrations, and ISRU of the Martian Moon as well as site reconnaissance for future human exploration of Mars.

GR.1	Mission must take place between 1/1/2015 and 12/31/2035
GR.2	Yearly NASA budget is $16 Billion (adjusting for inflation only)
GR.3	Must have a crew of four
GR.4	Must arrive at the surface of Phobos and/or Deimos
GR.5	Must stay on the surface of Phobos and/or Deimos for at least 300 days
GR.6	Must perform Mars moons surface exploration, technology demonstration, ISRU
GR.7	Must perform reconnaissance on Mars to facilitate future Mars human missions
GR.8	Must include tele-operated experiments on the surface of Mars
GR.9	Maintain at least 80% of NASA's total budget for existing NASA programs
GR.10	ISS will be fully funded until 2024
GR.11	SLS and Orion will be developed and operational through 2025 at their current budgets

Table 15. Ground Rules

		Reference
TL.1	Conduct a human mission to the moons of Mars between 1/1/2015 and 12/31/2035	GR.1
TL.2	Deliver and return four human crew members to /from the moons of Mars safely	GR.3, GR. 4
TL.3	Do not exceed a yearly NASA budget of $16 Billion adjusted for inflation and - Maintain at least 80% of NASA's total budget for existing NASA programs - ISS will be fully funded to 2024 - SLS and Orion will be developed and operational through 2025 at their current budgets	GR.2, GR.9, GR.10, GR.11
TL.4	Four crewmembers have to survive on moon surface an be able to conduct EVAs for at least 300 days	GR.5, GR.6
TL.5	Perform Mars moon surface exploration	GR.6
TL.6	Perform technology demonstration	GR.6
TL.7	Perform ISRU	GR.6
TL.8	Perform Mars reconnaissance	GR.7
TL.9	Prepare future human missions to Mars	GR.7
TL.10	Perform tele-operated experiments on the surface of Mars	GR.8

Table 16. Top Level Requirements

Appendix C: Risk Analysis and Mitigation Strategies

Risks related to all subsystems are rated according to the NASA risk management standard (NASA/SP-2011-3422).[6] The resulting risk matrix is shown in Figure 17. Mitigation strategies are implemented according to the severity of the risk and it is possible to reduce the majority of critical risks to a Loss of Mission (LOM) in the worst case, except for a failure of the crewed launch vehicle. The labels in the risk matrix refer to the numbering given to various risks and their respective mitigation strategies as listed below. Note that an inherent risk not shown in the matrix, but probably causing the mission to undergo major changes and cost increases is scheduling. This is due to a number of technologies that have to be developed from low TRL to at least TRL 6 or 7, and the required testing of critical technologies and launchers has to be considered. All of these developments need to be assessed critically and a rigorous timeline management needs to be implemented. Below is a list of the main mission risks along with their associated mitigation strategies. Their enumeration number corresponds to the number shown in the risk matrix (Figure 17)

Table 17. Risk matrix. Green, yellow and red stand for low, medium and high probability/consequence respectively. Rows = consequence; columns = probability.

	rare	unlikely	possible	likely	very likely
catastrophic	1,8,20,28	10			
major	11,13,17,30	19,26			
moderate	3,16,23,29,34	4,6,22,25,32,33	36		
minor	7,37	2,12,21,24,27	31,35	14	
insign.	15,18	5,38	9		

Trajectories

1. Science Payload: TMI maneuver is not fully successful. If the necessary delta-v to obtain the prescribed v-infinity cannot be achieved, this may result in a LOM for the scientific equipment. Another launch may be attempted resulting in a higher launch cost.

2. The lunar flyby maneuver during the inbound trajectory of the test crew vehicle is not timed correctly or fails. If Orions propulsion system is still working, a maneuver can be performed after the failed propulsive lunar flyby to return safely to Earth; TOF is estimated to be 6-10 days.

3. A subsystem such as ECLSS has a partial failure right after TLI and the test mission/main mission crew is required to be back at Earth as soon as possible. Mitigation A: if failure occurs within the first 3-4 days of TLI, a delta-v can be performed to change the outbound trajectory to a free-return trajectory. Estimated TOF from TLI to Earth reentry: 10-11 days. Mitigation B: if a failure occurs after 3-4 days from TLI, a delta-v can be performed at the lunar flyby to return to Earth safely without exceeding Orions reentry velocity capability.

4. Failed orbit insertion at H2. A propulsive maneuver can be performed at a later time than the nominal H2 insertion in order to arrive at a different halo orbit and then perform a rendezvous maneuver with HERMES+HARMONIA. Alternatively, if no alternative halo orbit can be achieved, perform a flyby of the Moon again and return safely back to Earth; TOF is estimated to be in the order of 8-13 days.

5. Maneuver to return to Earth at the end of the mission fails. A propulsive maneuver can be performed at a later time. This results in a small correction in order to return to Earth safely within 10 days and at nominal reentry velocity of 11 km/s.

Communications

6. Main communications system fails. Backup communication systems is used. Data rate may be lower.
7. Line of sight with Earth is obscured and communication with Earth is lost. Crew must wait until line of sight with Earth is reestablished.

Launch Vehicles

8. Falcon Heavy carrying the science mission malfunctions/fails to delive the payload into orbit. Enough buffer time is given between the science pre-deployment and the crewed mission so that another launch can be attempted. Results in higher cost and delay of science schedule.
9. Poor weather conditions do not allow the launch to occur on the nominal date. Reschedule the launch to a different date within the launch window.
10. SLS payload capacity is reduced. Perform the launch of HERMES and HARMONIA using two launches. Increased launch cost and may cause slight delay in launch schedule.
11. Falcon Heavy payload capacity is reduced. Margins ensure that the science mission may still be able to be launched using one Falcon Heavy. Otherwise, use 2 Falcon Heavy launches or decrease the amount of science equipment to be delivered at the Martian system.

Electrical Power System (EPS)

12. One SAFE-400 reactor fails. Less power can be delivered to the VASIMR engines, reducing thrust and increasing TOF. Stay time at Deimos may be shortened.
13. Two or more SAFE-400 reactors fail. LOM. Abort trajectory is implemented using the remaining power if possible. Otherwise, LOC.

Thermal Control System (TCS)

14. Unexpected eclipse from the Sun. Include at least one layer of MLI to ensure thermal inertia. Include heating device.
15. Coating absorptivity or emissivity degrades due to unexpected high solar radiation and/or galactic cosmic rays. Heating device and auxiliary radiator are utilized.
16. Heater/Radiator fails. If all radiators were to fail, crew may have to execute a premature Earth return.
17. Complete or partial system failure. It affects mainly EPS, causing a decreased power output and thus less thrust. Abort trajectory is implemented if necessary using the remaining power if possible. If failure is only minimal, stay time at Deimos may be decreased with no need for abort.

Environmental Control and Life Support System (ECLSS)

18. IVA suit failure. Use backup IVA suit.
19. EVA suit failure.EVA abort. Repair failure, use backup EVA suit, or use IVA suit in emergency case.
20. Cabin depressurization of either habitable vehicle. Launch: Abort mission (LOM), IVA suits will be donned and automatically pressurize and ensure crew safety until return to Earth. Transit: Enter other habitable vehicle and don IVA suits. Assess repairability and mission viability (May cause LOM). Reentry: Continue descent, IVA suits will be donned and automatically pressurize and ensure crew safety until return to Earth.
21. Torpor module failure. Awaken associated crewmember. Use spares to repair torpor module.
22. Sickness/injury of crewmember due to microgravity or torpor. Monitor crew health, follow mitigation techniques of known torpor risks, and follow proper workout protocol to reduce microgravity risks.

Further details concerning risk and mitigation strategies solely related to torpor can be found in Table 18.

Attitude and Orbit Control System (AOCS) and Landing/Ascent

Risk	Initiator	Mitigation Technique/Comments
Blood Clotting	Prolonged sleep and in-dwelling IVs	Minimize IV access, and perform periodic heparin flushed to dissolve clots
Bleeding	Decrease in coagulation factor activity	Not a significant concern outside of trauma
Infection	Temperature reduction in white blood cell activity	Minimize IV access, practice sterile techniques, and use of tunneled catheters and antibiotic-infused catheters
Electrolyte and Glucose Imbalances	Decreased cellular metabolism	Close monitoring of crew health and IV stabilization
Fatty Liver and Liver Failure	Long term torpor usage	Alternate source of lipids used, and proper diet and exercise when not in torpor
Other Complications	Torpor usage and reduced metabolic rate	Augment torpor system with insulin, exogenous CCK, and other risk-preventing hormones, and follow proper protocol for inducing and awaking from torpor

Table 18. Torpor Health Risks and Mitigation Strategies.[16]

23. AOCS thrusters underperform. Margins in propellant mass are taken into account to ensure the spacecraft has enough propellant should the AOCS thrusters underperform.

24. One or more AOCS thrusters malfunction and/or fail. Redundant/backup AOCS thrusters are used.

25. Landing gear does not function properly at landing or ascent. AOCS thrusters can be used as backup. May lower the science astronauts can perform at Deimos due to not being in direct contact with the surface of Deimos.

26. Docking with the resupply vehicle at Deimos fails. If no critical subsystems are damaged and enough delta-v is available, retry the docking maneuver; this may result in a reduced time for scientific exploration at Deimos. If docking with the resupply vehicle is impossible, the stay time at Deimos must be shortened to 100 days. Partial LOM.

Propulsion

27. One VASIMR engine fails. TOF is extended and stay time at Deimos is shortened. Two or more VASIMR engines fail. LOM. Abort trajectory is implemented using the remaining engines if possible. Otherwise, LOC.

28. Fuel leakage caused by micrometeorite impacts. Crew may be able to repair the damage by going outside using EVA suits. If the damage cannot be repaired, mission is aborted causing LOM.

Radiation Shielding

30. No adequate shielding material is developed/researched for the main mission timeframe. Allocate more research funds to the shielding material. May cause delays and/or LOM.

31. Underestimated length of radiation event. Astronaut schedule may be changed to accommodate to the unexpected/underestimated radiation event.

Robotics

32. Moon hoppers get stuck in the Martian moon's terrain. Astronauts can try to teleoperate the moon hoppers to get them unstuck.

33. Moon hoppers are covered in dust and do not receive enough solar energy from their solar arrays. Redundancy and margins. Science return may be diminished.

34. Springs mounted on the moon hoppers used for mobility malfunction. Loss of moon hopper. Redundancy assures that another moon hopper would be available.

Science

35. One or more MAN stations malfunction. Network covered by the MAN stations is reduced. The high number of MAN stations deployed provides redundancy.

36. Space Solar Power Station does not deliver enough power to all the MAN stations. Some MAN stations may not be able to function reducing the coverage of the MAN station network.

37. ISRU equipment does not function properly/malfunctions. ISRU experiments may not be conducted as intended. Lower science return. The crew is not affected.

38. Communication between astronauts and equipment on the Martian surface partially/completely malfunctions. Backup communication systems are used.

Appendix D: MAN Station Mass Breakdown

Tables 19, 20, and 21 provide a detailed summary of the mass breakdown for each portion of the MAN stations: scientific payload, lander, probe respectively.

Scientific Payload	**Mass [kg]**
Seismometer and ground heat probe	3
Temperature, wind and humidity sensor	2
Radiation sensor	0.5
360 degree panoramic camera	0.5
Soil test instrument	1
Organics test instrument	5
Microscope imager to determine regolith grain size	0.5
Subtotal	12.5

Table 19. Scientific Payload Mass Budget

Lander	**Mass [kg]**
Structure	12
Microwave receiver	1
Antenna for data transmission	1
Miscellaneous (battery, electronics, cabling etc)	10
Subtotal	24

Table 20. Lander Mass Budget

Probe	**Mass [kg]**
Structure (heatshield & back cover)	18
Parachutes	3
Airbags & gas generator	14
Subtotal	35

Table 21. Probe Mass Budget

Appendix E: Low-Thrust Trajectory Optimization

The optimal low-thrust interplanetary trajectory from the SOI of the Earth to the SOI of Mars has been computed considering the real ephemerides of Earth and Mars at given departure and arrival dates.[32]

Electric propulsion, while highly efficient, requires the engines to operate during a significant fraction of the trajectory and this makes it particularly difficult to find optimal trajectories.[33] The methods used to solve the low-thrust trajectory optimization problem generally fall into two categories: direct and indirect methods. Indirect methods are based on calculus of variations and on the formulation of a two-point boundary problem involving a set of costate variables, the solution of which yields a history of the time-dependent controls. Finding a solution using indirect method is often difficult because of several reasons: the size of the dynamical system doubles in size when adding the costate variables, the convergence domain tends to be small and the problem is sensitive to the initial guesses of the costate variables, which are generally not physically intuitive. Direct methods, on the other hand, are based on the parametrization of the controls and use nonlinear programming (NLP) techniques to optimise the performance index. Advantages of direct methods are the increased computational efficiency, more robust convergence and a reduced sensitivity to the initial guess, which is moreover physically more intuitive than for indirect methods. Different methods are available to solve direct optimization method, e.g., single shooting, multiple shooting and collocation.

The optimal low-thrust trajectory for the transfer from Earth to Mars has been computed using a direct method and a multiple shooting algorithm. The trajectory is segmented into a sequence of coast and thrust legs. The objective of the non linear programming problem is to minimize the propellant consumption subjects to constraints (the initial state vector of the spacecraft has to coincide with the state vector of the Earth at departure, the final state vector has to coincide with the state vector of Mars at arrival, the initial and final points of the coast and thrust legs have to match). The non-linear programming problem has been solved using the Matlab® *fmincon-interior point* algorithm. The variables to optimize are the state vectors at the initial and final point of each thrust legs and the thrust direction over those legs.

The model used by the optimization method is an analytical propagator for the trajectory subject to the low-thrust acceleration.[34] This speeds up the computational process with respect to a numerical propagation, since in an optimization problem the trajectory has to be evaluated several time.

Appendix F: Resupply interplanetary trajectory

The method used to compute the resupply interplanetary trajectory for the Martian Moons Resupply and Science Deployment (MMRSD) is described in Appendix E. The obtained trajectory for this resupply and science deployment is shown in Figure 14, with thrusting arcs shown in red and coasting arcs in green. The circles along the trajectory show points where the thrust angle direction is changed for the next thrust arc.

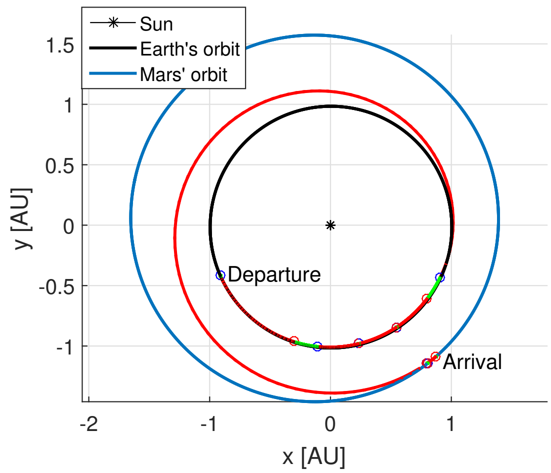

Figure 14. Interplanetary trajectory from the SOI of Earth to the SOI of Mars for MMRSD

Appendix G: Mothership Diagram and Team Picture

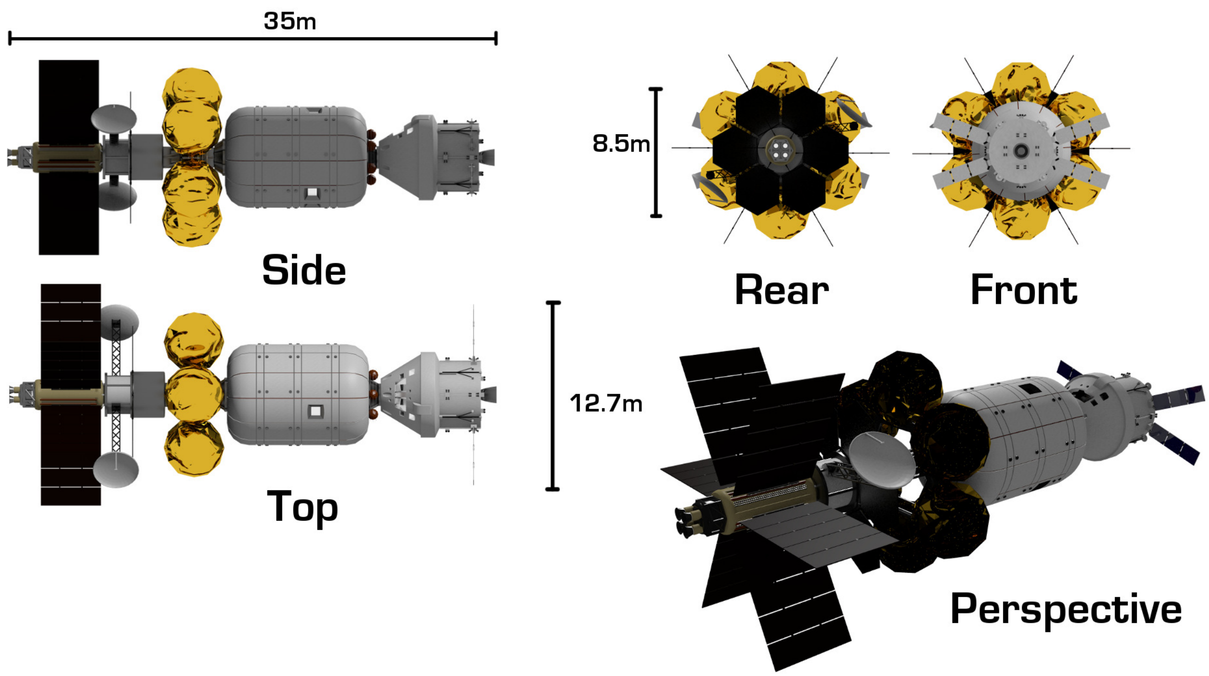

Figure 15. Mothership diagram

Figure 16. Team Picture