

LION: laser interferometer on the moon

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Received 18 December 2020, revised 30 March 2021

Accepted for publication 1 April 2021

Published 24 May 2021



CrossMark

Abstract

Gravitational wave astronomy has now left its infancy and has become an important tool for probing the most violent phenomena in our Universe. The LIGO/Virgo-KAGRA collaboration operates ground based detectors which cover the frequency band from 10 Hz to the kHz regime. Meanwhile, the pulsar timing array and the soon to launch LISA mission will cover frequencies below 0.1 Hz, leaving a gap in detectable gravitational wave frequencies. Here we show how a laser interferometer on the moon (LION) gravitational wave detector would be sensitive to frequencies from sub Hz to kHz. We find that the sensitivity curve is such that LION can measure compact binaries with masses between 10 and $100M_{\odot}$ at cosmological distances, with redshifts as high as $z = 100$ and beyond, depending on the spin and the mass ratio of the binaries. LION can detect binaries of compact objects with higher-masses, with very large signal-to-noise ratios (SNRs), help us to understand how supermassive

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black holes got their colossal masses on the cosmological landscape, and it can observe in detail intermediate-mass ratio inspirals at distances as large as at least 100 Gpc. Compact binaries that never reach the LIGO/Virgo sensitivity band can spend significant amounts of time in the LION band, while sources present in the LISA band can be picked up by the detector and observed until their final merger. Since LION covers the deci-Hertz regime with such large SNRs, it truly achieves the dream of multi messenger astronomy.

Keywords: gravitational wave detector, lunar, detector concepts, deci-Hertz

(Some figures may appear in colour only in the online journal)

1. Introduction

The LIGO-Virgo-KAGRA (LVK) collaboration [1–3] now forms a network of gravitational wave observatories covering the Earth. Third generation ground based detectors such as the Einstein telescope (ET) [4] and cosmic explorer (CE) [5, 6] are currently in the planning stage of development. These detectors are planned to push the lower frequency bound of Earth based gravitational wave detectors to their fundamental limits.

Newtonian noise, or gravitational gradient noise, forms a lower frequency bound of Earth based detection at about 1 Hz [7]. Meanwhile, with a proposed launch of 2034, the ESA’s LISA mission [8] will cover the μHz to sub deci-Hz regime of the gravitational wave spectrum. The upper sensitivity bound of space based detectors is set by the interferometer arm length [9, 10].

These barriers create a gap of measurable frequencies which requires a different style of mission to fill. In this gap many interesting stellar phenomena go unrecorded and their immeasurable impact on the field of gravitational wave astronomy is lost [11]. Therefore, there is great interest in any gravitational wave detector capable of bridging this gap. An ambitious mission proposed by the Japanese gravitational community called DECIGO is in the planning stage of development [12]. DECIGO aims to cover the deci-Hertz band; however, it is a new style of detector, which requires many technological innovations before it can launch [13, 14]. This mission, and others [15], are still a long way from realisation, so in order to cover these frequencies in the operational time of third generation detectors, other avenues of low frequency detection must be explored. In this work we show that a lunar gravitational wave observatory would be able to observe frequencies down to 0.7 Hz while simultaneously overlapping with the sensitive frequency range of Earth based detectors.

During the 90s Wilson and La Fave published two papers on the idea of lunar interferometers, but as the field was still in its infancy, a lot of benefits and challenges went overlooked [16, 17]. A few other groups have suggested using seismometer arrays on both the Moon and other extraterrestrial solar bodies [18, 19]. Coughlin and Harms managed to put limits on the energy contained within the stochastic background across the frequency band 0.1–1 Hz using seismometers on the Moon from the Apollo missions as a detector. A recent proposal has now been submitted to launch many high sensitivity seismometers to the surface of the Moon to push this detection method to its limits [20]. If launched, these seismometers would enable a better understanding of the Moon’s seismic activity and so the true limits of seismic noise as discussed later in this paper. Aside from that, a paper has been submitted to arxiv recently discussing the concept of a lunar gravitational wave detector [21]. They focus largely on the astrophysical sources in their sensitivity band. Although their optimistic estimate of performance was excellent at low frequency, it relied upon the use of a freely floating test mass, a technology that has only been proposed and has no further development so far [22]. The

Table 1. Overview of the key parameters of LION. The masses and suspension length are listed from test mass to top mass. The model is based on the gravitational wave interferometer noise calculator (gwinc) CE 2 from October 2020. The full model and adapted gwinc software is provided as supplementary material. The full detector configuration files can be found at <https://github.com/Jonjocarts/LION-Public>.

Parameter	Value
Laser power	50 W
Wavelength	2000 nm
Arm length	40 km
Seismic	Earth surface/1000
Test mass	1267/1267/698/707 kg
ITM radius of curvature	34 km
ETM radius of curvature	36 km
Suspension length	3.54/2.05/1.66/2.50 m
Temperature	70 K
Signal recycling cavity length	55 m
Squeezing	
Initial squeezing level	15 dB
Filter cavity length	8.94 km
Filter cavity detuning	−2.3 Hz
Mirror transmittance	
End test mass	5 ppm
Input test mass	1.2%
Power recycling mirror	3%
Signal recycling mirror	2%
Filter cavity input mirror	0.17%
Filter cavity end mirror	5 ppm

conservative estimate they make is similar to the one we reach here, without focusing on the technical details needed to reach that. We also show significant differences in detector design. Still, it is very promising that two independent approaches can reach similar noise estimates.

In the following we discuss how a ground based long arm interferometer with Fabry–Pérot cavities could be constructed for use as a gravitational wave detector on the surface of the Moon. We will discuss the merits and potential design style of such a lunar observatory. We address this by constructing a feasible noise budget for a laser observatory based upon current gravitational wave detectors. We benefit from the lunar surface’s seismically quiet environment [23], and its lack of atmosphere. Thus the limit on low frequency detection set by the Newtonian noise can be avoided. By means of this noise budget, we show that the sensitivity gap between the high frequency ground based detectors and low frequency space based detectors can be partially bridged. Furthermore, a detector built on the Moon with a sensitivity band also overlapping that of the Earth based network increases the maximal distance between the detectors by a factor of 30, improving the source localisation ability significantly.

Several stellar phenomena are expected in the frequency band exclusively covered by our proposed lunar detector and we discuss the impact their observations will have. A low frequency detector allows the tracking of merging binaries for much longer times [24], detecting binaries at much greater redshift [25], detecting heavier binaries [26], allows for better sky localisation [27], and detecting new types of astrophysical sources [28, 29]. Long standing questions about the origins of many sources could uniquely be resolved with a LION detector [30]. Furthermore, sub-Hertz detection allows for the probing of gravity in the strong field

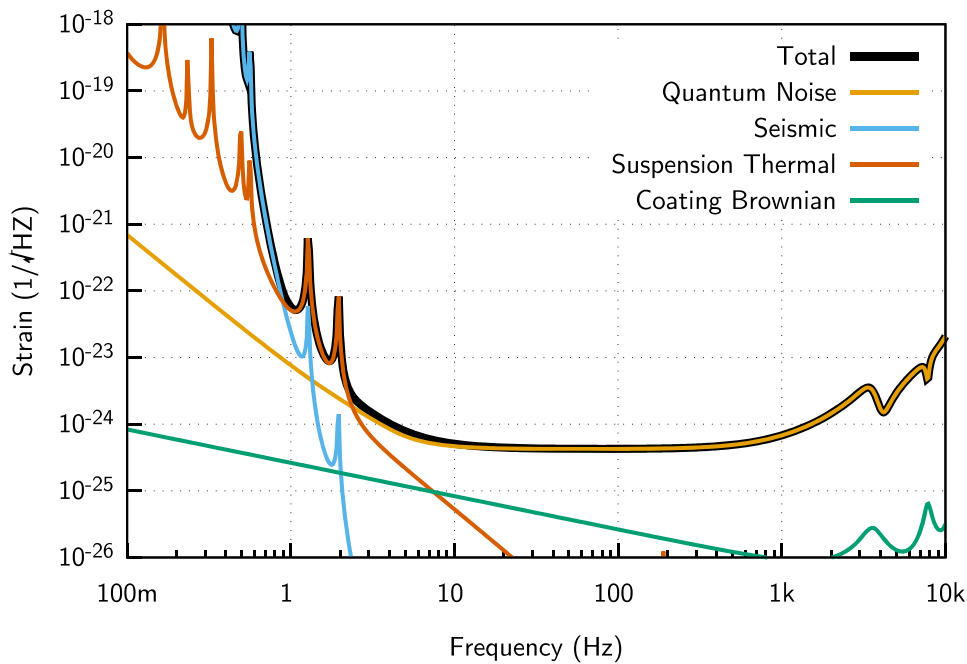


Figure 1. Noise budget for a gravitational wave detector operated on the Moon with an arm length of 40 km and optics with increased masses at a temperature of 70 K. The parameters used to create this are listed in table 1. The detector is quantum noise limited above 2 Hz and limited by thermal noise below this until it hits the seismic cliff at 0.8 Hz. The bandwidth of the detector is therefore approximately from 0.7 Hz to the kHz. Simulated with gwinc [38].

regime, testing general relativity to its limits [31–33]. Finding sources at very high redshifts leads directly to a better understanding of the early Universe, where our current understanding is limited by our detection capabilities.

With the renewed interest in lunar missions from several major global powers [34–37], now is the ideal time for the gravitational wave community to begin considering what use the Moon can be in aiding our understanding of the cosmos.

Section 2 starts with the discussion of technical details, including detector parameters, noise contributions and the location and infrastructure. This is followed in section 3 by a summarised noise budget. The science case in section 4 discusses the new possibilities that a lunar detector offers, while section 5 gives an estimate about the mission cost and timeline.

2. Technical discussion

The model for our suggested laser interferometer on the moon (LION) is based on the CE 2 proposal. The key parameters are listed in table 1. Our changes arise from the special conditions on the Moon and design choices made to optimise the frequency band of interest which we discuss below. This leads to the noise curves shown in figure 1. The overall schematic is shown in figure 2.

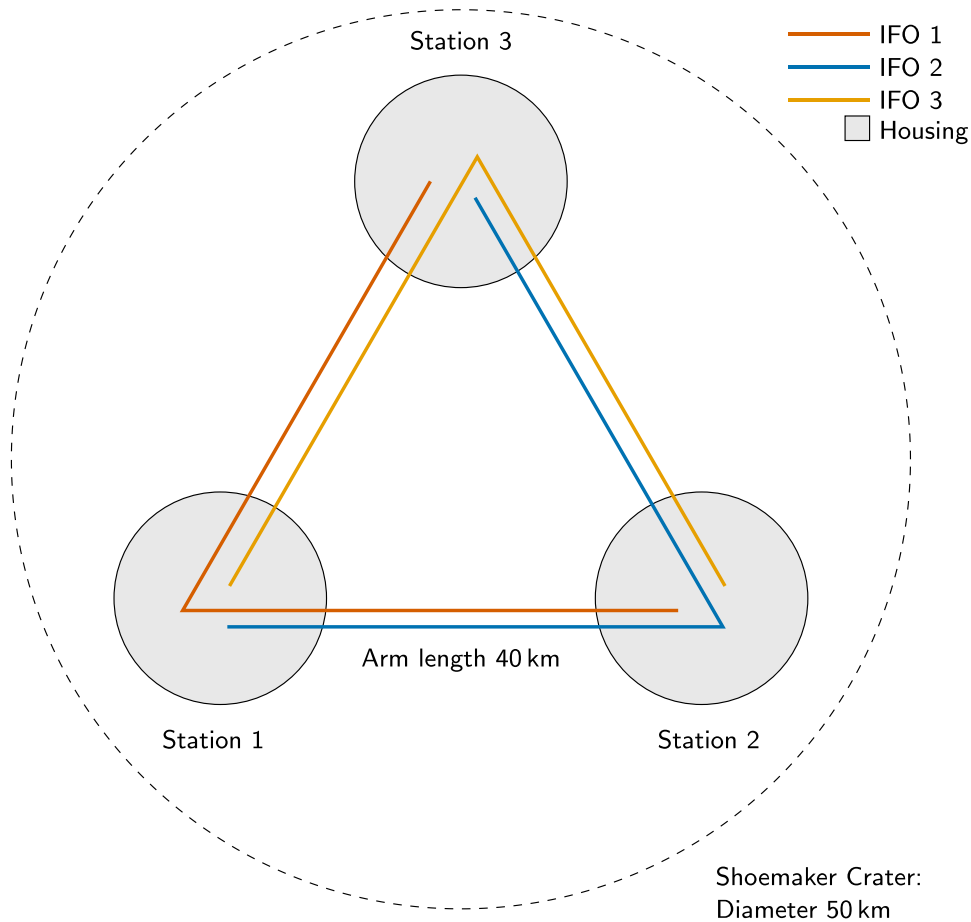


Figure 2. A schematic representation of the LION detector, showing the three interferometers and end test stations. The arms are all 40 km in length. The end stations must be protected from dust entering them and each contain isolation platforms and suspension systems for the core and auxiliary optics.

2.1. Location

The choice of the location for LION is predominately determined by the temperature map of the Moon. A crater near a lunar pole is considered by us to be a suitable position for several reasons; thermal stability, geometry, and shelter from solar radiation and dust [39–41]. A promising candidate site is the Shoemaker crater (88.1°S 44.9°E) at the Moon’s south pole. Hayne *et al* [39] investigated the measurement data of the lunar reconnaissance orbiter (LRO) mission launched in 2009 for three craters with about 50 km diameter. The shoemaker crater shows an average temperature of about 50 K with a relatively small variation of minimal/maximal temperature 20 – 95 K compared to other parts of the Moon. It has a diameter of 50 km, introducing one of the limiting factors for the detectors: the arm length. The nearby Malapert mountain is a nearly perfect site to build solar panels and a communication equipment for data transfer [42]. Additionally, the surface appears to be nearly free of water frost, which

could otherwise evaporate onto optics [39]. We assume the use of this crater for all future calculations.

2.2. Infrastructure

The vacuum on the surface of the Moon is three orders of magnitude better than that in the vacuum tubes in the aLIGO detectors [1, 43], and so no infrastructure will be built to house a vacuum. Only the optic support stations are needed to isolate the test optics and allow for their precise alignment controls.

For the detection of gravitational waves, coincidence data from two independent simultaneous instruments is required. This is solved in the ET detector and LISA by using a triangular shape with three interferometers in the arms [4, 44, 45]. Therefore, we propose to build LION in a triangular shape with an arm length of 40 km, limited by the dimensions of the crater. The three interferometers of LION will also allow for the detection of gravitational waves regardless of polarisation, as shown in figure 2.

2.3. Background seismic

The seismic noise on the Moon arises from two main sources; meteorite impacts [46–49] and from the shallow seismic activity of the Moon [50]. Many of the sources of seismic activity on Earth do not exist on the Moon [51] such as moving cars and trains, winds, and the oceans tides. In particular, the oceans tides cause a large microseismic peak at 0.1–0.2 Hz which proves troublesome to all detectors seeking low frequency performance; a lunar site completely negates this [52]. Meanwhile, wind and ground conditions on the Earth tilts the detectors, polluting the measurements of inertial sensors with additional noise, which is directly injected into platform controls [53, 54]. The absence, or at the very least substantial reduction, of tilt would allow for a much more aggressive active platform control scheme to reduce motion in the test masses [55]. The control noise is currently one of the reasons the aLIGO detectors do not reach their design sensitivity at low frequencies.

The only direct seismic measurements on the Moon’s surface were performed during the Apollo missions [23, 46]. They concluded that the background seismic level was significantly below the sensitivity of their measurement devices ($0.3 \text{ nm Hz}^{-1/2}$ at 1 Hz), and thus lower than anything on Earth’s surface. From this we can safely assume the seismic on the Moon to be smaller by a factor of 1000 than Earth’s surface seismic motion. However, the Russian operated Luna 27 will be launched in 2024 [56] with a more sensitive seismometer [57], possibly measuring an even lower boundary.

2.4. Newtonian noise

Newtonian noise is caused by the movement of mass around the detector causing a change in the local gravitational field [51]. These largely come from two sources; the atmosphere and the ground. The lack of atmosphere on the Moon negates the former entirely. Reliable modeling of seismic Newtonian noise for the lunar surface is impossible without accurate measurements of the seismic activity and surface composition. On Earth the noise forms a noise cliff at approximately 1 Hz. Since the Moon has lower activity [47] and has a lower density on its surface it is assumed that this will occur at a lower frequency. Newtonian noise is therefore not plotted in the noise budget. A full study of the gradient noise would still be essential as more knowledge of the Moon’s activity is gained.

2.5. Suspension systems

The limiting low frequency noise considerations are the residual of seismic activity and suspension thermal noise. Like in Earth bound detectors, a multistage seismic isolation system would be necessary. This will comprise of a preisolation stage and several suspended masses each acting as a pendulum [58, 59]. In order to achieve the required seismic isolation, the resonance frequencies of the pendula must be tuned to lower frequency than CE. The gravity on the Moon's surface is 1.62 m s^{-2} , approximately a sixth of the gravity on Earth. Although the lower gravity of the Moon offers lower natural frequency by a factor of 0.4, longer suspensions will still be needed. Furthermore, to help combat thermal noise, heavier test masses must still be included. As the gravity is weaker, the masses can be a factor of 6 larger than CE without additional weight on the suspensions. Making such large test masses and coating them may prove difficult so we assume only a factor of 4 increase can be made. This leads to a test and penultimate mass of 1270 kg and another 2 stages with about 700 kg mass. Making the lengths of suspensions on average a factor of 3 longer allows for the whole system to fit in a 11 m structure while reducing the noise to a level suitable for sub Hz measurements. Suspension systems are a limiting factor for the low frequency performance of all detectors with fundamental problems of how much weight can be suspended and resonance frequency. The lower gravity on the Moon provides help on both of these fronts. How far this can be developed will form the lower boundary of measurable frequencies, but ultimately it is a matter of engineering.

2.6. Thermal noise

Thermal noise is a considerable problem for the low frequency response of any detector. The extremely low temperatures at the Moon's south pole are a good start for achieving the necessary thermal noise suppression. However, a low mechanical loss is also a necessity for a low thermal noise [60]. The bulk material of the test masses at both LIGO and Virgo are fused silica which, due to a mechanical loss peak, has relatively high thermal noise at cryogenic temperatures [61]. Research for alternative test mass materials is ongoing for use in the cryogenic interferometer of ET and other detectors [62–64].

The CE proposal uses silicon ribbons to suspend their test masses and research is ongoing to design the next generation of suspensions [65]. The properties of the ribbons quoted in the CE design is assumed when calculating the suspension thermal noise.

Likewise, significant progress is needed to reduce both mechanical and optical losses of the test mass coatings to reach CE's design sensitivity [6]. The progress of these development will be at the forefront of the efforts of next generation of gravitational wave detectors and their results will heavily influence the limits of a lunar interferometer. The LION design does not need to exceed the performance of CE in this regard so it is assumed that if CE can meet the design sensitivity, the design can be copied for LION. Promising results in this regards are already being achieved by two means of coating; one with alternate material layer deposition [66] and the other by the structuring of resonant waveguides to create a highly reflective surface [67, 68].

2.7. Quantum noise

Quantum noise is the predominant noise source at high frequencies for Earth bound interferometric gravitational wave detectors. While frequency independent squeezing is injected in current detectors [69–71] to reduce the shot noise, there are plans for a broadband noise reduction by using filter cavities [72] or EPR entanglement [73]. The proposed LION design uses 50 W laser power, 15 dB initial squeezing and filter cavities. Compared to the CE design the

power was reduced by a factor of 5 to increase the sensitivity at low frequencies. The detector will be quantum noise limited above 2 Hz.

2.8. Seismic shocks and asteroid impacts

Another cause of concern is asteroid impacts on the Moon. A large impact near the equipment could potentially lead both to saturated isolation equipment leading to damage to the controls as well as large mirror motion and loss of science mode. Reacquiring science mode is a difficult and time consuming process [74, 75], however, a similar problem is encountered on Earth with earthquakes. This problem is tackled at aLIGO with a combination of a specialised early warning system for incoming high activity seismic waves and alternative control schemes for high activity times [76]. A similar strategy could be copied for any lunar setup. Therefore, only a direct hit on the experiment would present a problem.

Based on the data of the still ongoing LRO, approximately 47 000 new impact sites had been detected in the area studied (6% of the lunar surface) in a 1241 day period [77]. This leads to an estimate of approximately 5 impacts within the detector area (a triangle with 40 km long sides) in a year. It is foreseen that this will prove to be a regular nuisance, but should still enable long stretches of observation. It is also unlikely that the optics will be hit directly as they will take up a small fractional area of the detector, leading to an estimate of direct hits once in a thousand years. Smaller impacts will be significantly under-counted by their very nature, and so it is likely a asteroid shielding solution of the optics will be needed.

2.9. Dust accumulation

The Apollo missions showed that the Moon's surface is covered in dust, which is kicked up by both activity, such as asteroid impacts and moonquakes, and solar winds [78, 79] and then coats everything on the lunar surface. This dust presents a very real challenge to the LION mission as all current gravitational wave detectors operate in the highest level of clean room environment. Dust stuck to the mirrors would act as a loss channel leading to an increase in quantum noise [80]. The increase in noise corresponding to different optical losses is shown in figure 3. To be a useful contribution to a gravitational wave network, optical losses at the mirror or along the path must be maintained below 0.1%. In addition, each dust particle will act as a point absorber on the mirror. The LIGO detectors have discovered that these point absorbers exist in their mirrors, which absorb more of the light until they are heated enough to damage the mirrors [81]. It is, therefore, imperative that the mirrors remain as immaculate as possible. Due again to the need to suppress thermal noise, the material suspending the optics must also remain protected from dust. Any dust which does accumulate will degrade the quality factor of the suspensions and so lead to worse performance.

Dust does present a problem for many experiments on the Moon and other bodies in the Solar System; hence, much research is ongoing to combat this [82–91]. For LION, two dust reduction methods are promising, yet. Firstly, we mitigate the dust to stuck on our instruments by building everything in a closed housing, only accessible for the beam by a hole in direction of the beam path. The diameter of the hole must be at least three times the diameter of the beam to avoid a reduction of power circulating in the cavity. For a beam diameter of 360 mm this would give a hole diameter of about a meter. To prevent the dust from entering the housing via this hole, we consider adding a tunnel of a few meters length in front of the hole. This solution would be the simplest mechanically and cost wise. It may be the case that, after testing, it is found that further measures are required. To this end the second proposed mitigation technique for LION is an electrostatic dust shield at the instrument housings, which is currently considered to be the state of the art for cleaning optical elements [92]. Here, electrodes printed on a glass substrate

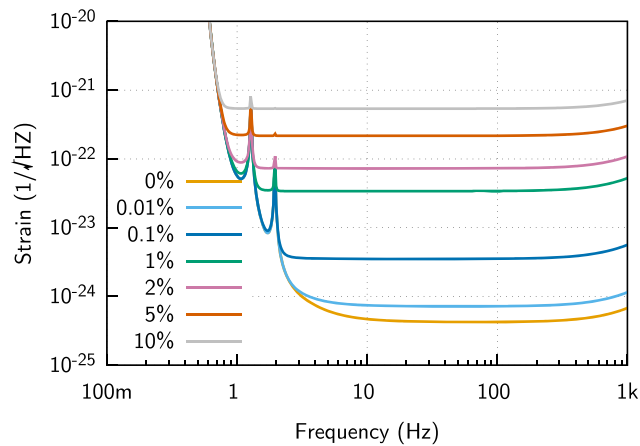


Figure 3. The effect of additional optical losses from dust on the mirror on the noise of the detector. The losses refer to optical power loss, per one travel of the cavity. Above a loss of 0.1% the detector makes few contributions to any proposed network and so a means of cleaning to this amount must be found. Losses highly influence, the frequency response of the interferometer and thus the filter cavity design. Therefore, we have assumed an optimised squeezing angle for this calculations, which artificially improves the low-frequency behavior of the 0% and 0.1% curves.

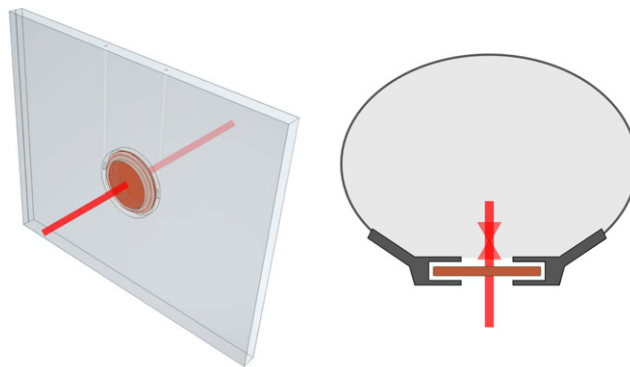


Figure 4. Schematic three dimensional view (left) and top view (right) of the dust protection method at the end stations, where an optic at the stations functions as an electrostatic dust shield. Since the optic has to be suspended, the station cannot be enclosed completely. Therefore, the wall is constructed around the window with tubes for the suspension, as shown in the left picture. The top view on the right side shows the resulting, very small space, that remains between the housing and the optic. The optical system including the test masses are placed behind this window and are not shown in this sketch.

move the electrostatic field toward the outer part of the optic. When utilised, the electrostatic wave removes both the uncharged particles as well as dust particles charged by the irradiation of solar winds and cosmic rays. Kawamoto *et al* have shown that electrostatic cleaning can limit the transparency loss to below 9% in experiments [87, 88]. This will likely be improved by the orientation of the optics in LION which are placed orthogonal to the lunar surface. Hence no gravitational forces will hold the dust particles at their surface. Further research on

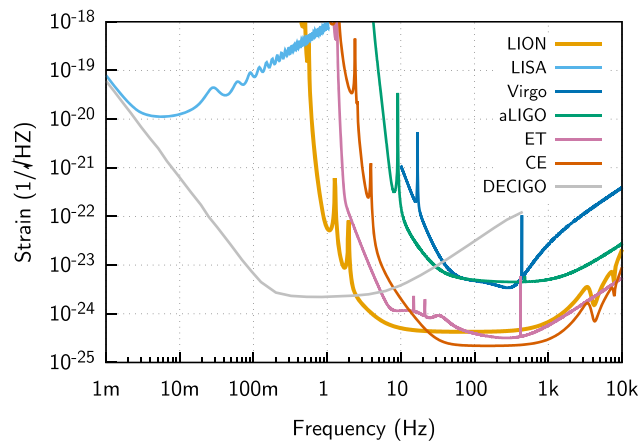


Figure 5. Comparison of the design sensitive of current and future gravitational wave detectors. Our proposed detector, LION, could extend the range of ground based detectors reducing the gap of measurable frequencies between Earth based and space-borne missions. LION has the highest sensitivity in the detection band from below 30 Hz and is the only detector able to with sufficient sensitivity for detection between 0.7 Hz and 2 Hz.

this topic will show if this technique can be refined to a sufficient level. If an electrostatic dust shield were to be implemented, it would have to be suspended to isolate it from environmental conditions. A schematic view of this is shown in figure 4.

2.10. Additional challenges

A mission of this style will require several large payloads to be transported to the Moon and positioned carefully, requiring assembly either by hand or robot. The feasibility of these strongly depends directly on the lunar missions in the coming decade. However, technology is reaching the stage where commercial missions to the Moon are becoming viable, a promising sign of things to come [93].

The total mass of the suspended optics is 4000 kg, and the weight of a preisolation stage used at LIGO is 2000 kg [52]. This would leave over 10 000 kg as remaining load for both the structure and auxiliary optics, if it was to be carried by the highest capacity rocket today. This would require at least 3 launches for total transport.

Gravitational wave detectors produce a high amount of data due to the number of sensors monitoring the numerous degrees of freedom. Furthermore, many systems will have to be operated from ground stations. A high bandwidth connection will be required to the site. Currently a record of 622 Megabits/s data transfer rate has been achieved by the lunar laser communication demonstration in orbit of the Moon [94]. With an encoding of 64 bits per data point, we would be able to read out about 100 channels at a 100 kHz sample rate. This 100 kHz rate is an upper estimate of the data rate needed. While this may not encompass all of the data producing instruments, it should be sufficient for instrument readout of the main channel and diagnostics of auxiliary channels of interest to above the upper limits of LION's sensitive band.

With no need for a vacuum pump, and only small amounts of power needed for alignment controls and isolation, the main power consumption concerns are the telemetry, the high laser power, beam preparation and readout electronics, and the auxiliary sensors. In space missions the power supply is usually covered by a combination of advanced solar panels that reach an

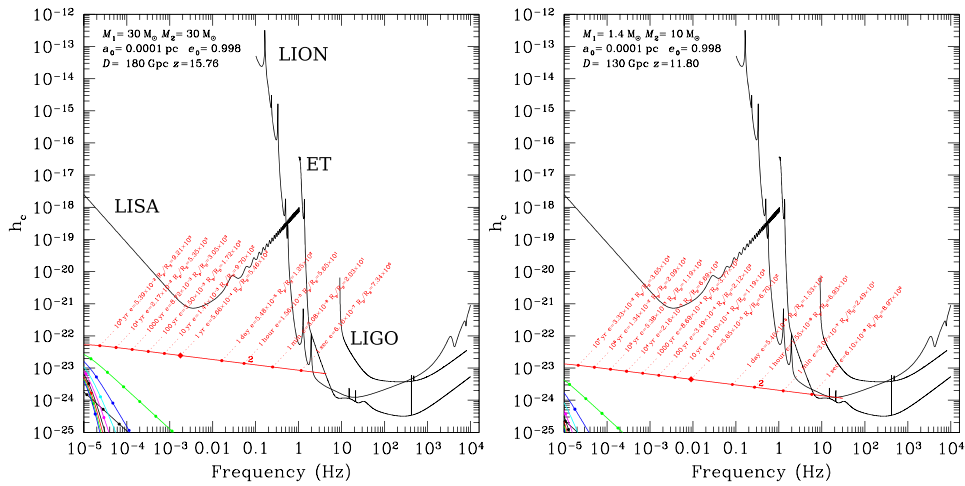


Figure 6. Evolution in frequency of the first harmonics (displayed in different colors) of two compact binaries in the approximation of Keplerian ellipses of [99]. We depict the characteristic, dimensionless amplitude as a function of the frequency in Hz. The initial dynamical parameters corresponding to the each source is summarised at the top, left corner of each panel (masses, initial semi-major axis and eccentricity and distance to the source D). From the left (lower frequencies) to the right (higher frequencies), we show the sensitivity curve of LISA [8], then LION, starting at about 1 Hz, the ET [100], and then LIGO. The upper, red harmonic corresponds to $n = 2$ in equation (2), which means that the source has circularised. We also show different moments in the evolution of the harmonics, and add the information relative to the source at that moment; in particular the remaining time until merger, the eccentricity and the periastron distance R_p in Schwarzschild radii R_S . The left panel shows a binary of masses $M_1 = M_2 = 30M_\odot$ at a distance of 180 Gpc, which corresponds to a redshift of $z = 15.76$. This is orders of magnitude distance further away than what LISA can hope to observe. The right panel shows the inspiral of a compact object of mass $1.4M_\odot$ (e.g. a neutron star) onto a stellar-mass black hole of mass $10M_\odot$ at a distance of 130 Gpc ($z = 11.80$). Even at this cosmological distance, the source spends one second in the LION domain. We note that no other detector is in the position of resolving these sources.

efficiency of 30% and Li-Ion batteries. While at the equatorial and mid-latitude regions of the Moon the long lunar nights set limitations for solar power supply, we take advantage of locating the interferometer in polar regions. As shown by [42] we receive up to 93% of full or partial sunlight when placing the solar panels on mountains, for example the Malapert mountain, close to the chosen site. For 1.4 kW m^{-2} of power provided by the Sun and a pessimistic factor of 0.5 for the partial sunlight, we gain 0.2 kW m^{-2} of solar power, making the power of the system very manageable with a solar panel array of a few tens of square meters.

3. Noise budget

We used the gwinc [38] software to estimate the noise budget of our proposed LION design shown in figure 1. Cosmic Explorer (CE2 model, October 2020 [6]) was used as a base model, but with the changes discussed in the previous section and listed in table 1 to match the design presented in this paper. The model is currently limited at low frequencies by the properties of the suspension system. Figure 5 shows the LION noise budget estimate and compares it with the performance of other current and proposed ground based detectors and the space

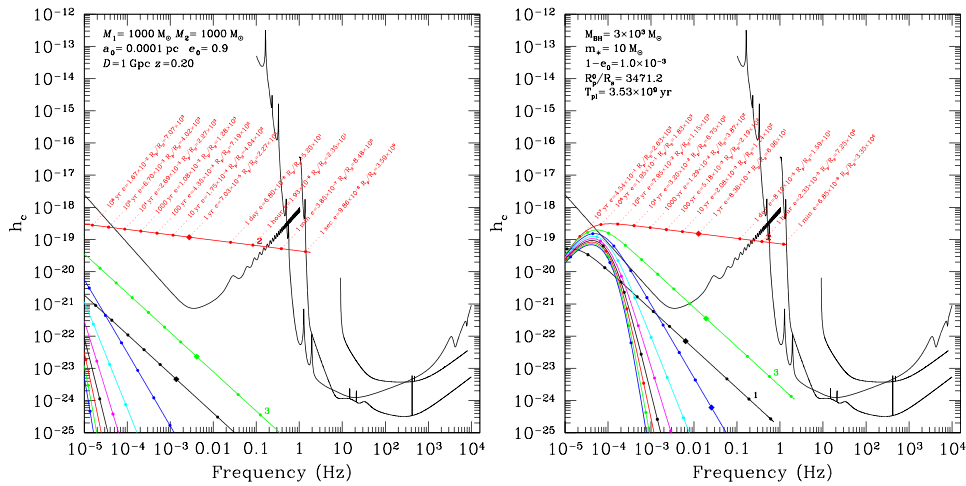


Figure 7. Same as in figure 6 but for a binary of two IMBH (left panel) and an IMRI composed of an intermediate-mass black hole of mass $M_{\text{BH}} = 3000M_{\odot}$ and a stellar-mass one of $m_* = 10M_{\odot}$ (right panel). In the latter we also include information about the initial periastron distance R_p^0 , as well as the initial time for merger T_{pl} .

based detector LISA. Although our estimate is based upon current technology and we possibly overestimate the Moon's seismic activity, LION performs better than ET and CE at frequencies below 30 Hz with detection capabilities down to 0.7 Hz. LION also manages to maintain performance comparable to CE into the kHz regime.

4. Science case

The LION sensitive band, from figure 5, ranges between 0.7 Hz and several kHz with peak sensitivity at 10 Hz. Several investigations of sources in this frequency range have been conducted [95], showing the great interest in this band, a region where the current ground based network is blind to. In general, detecting in a lower frequency range opens the possibility for observing more massive black holes and thus allows to study gravity in the strong field regime [29]. This is a key element in the testing of general relativity in more extreme limits. Many new potential sources are predicted to exist in this band, for example, type 1A supernovae, which have a gravitational wave frequency signature around 1 Hz. As discussed in [28], their observation may resolve the open question of their origins.

What makes LION unique, however, is its ability to cover a wide regime of frequencies and characteristic strains of merging binaries which cannot be explored with any other observatory. This forms the focus for this section.

4.1. An approximation

We first give a few examples approximating the evolution in phase-space of the harmonics of the characteristic strain with the method of Peters and Mathews [96]. This has the advantage of allowing us to explore the evolution of the eccentricity, which is rather limited in more realistic waveforms based on computations of full general relativity. Once we have an idea of the evolution in phase-space of these systems, we will investigate in more detail the evolution of the full waveform and the computation of the signal-to-noise ratio (SNR) following the

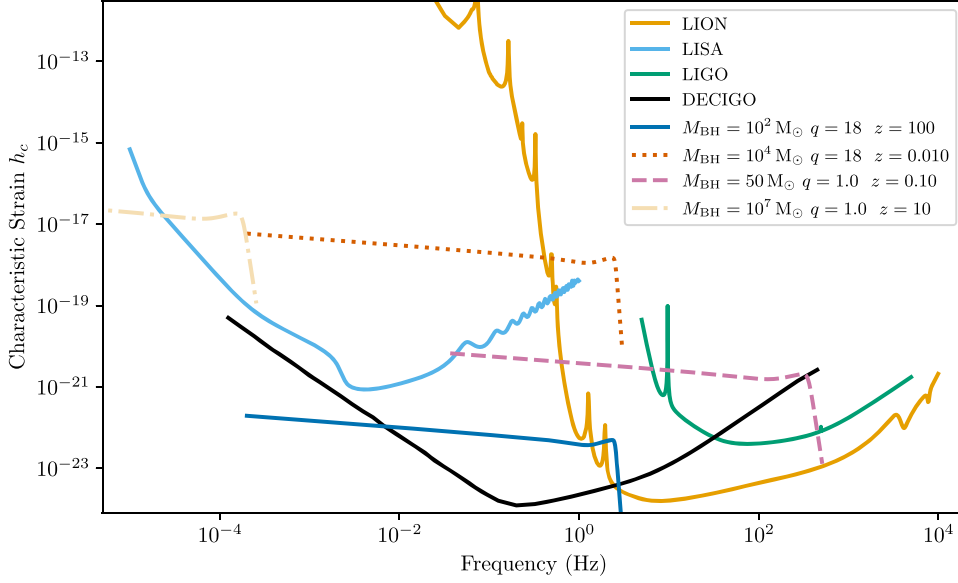


Figure 8. Evolution of four different signals with masses ranging between $10^7 M_\odot$ and $50 M_\odot$, different mass ratios q (in the legend we show the mass of the largest black hole) and the assumed redshift. We show three different observatories (LISA, LION and LIGO). The yellow, dash-dotted waveform corresponds to a system of $10^7 M_\odot$ at a redshift $z = 10$ with $q = 1$; the orange, solid waveform to a binary of $q = 100 M_\odot$, $z = 100$ and $q = 18$; the blue, dotted one to a system of $10^4 M_\odot$, $z = 10^{-2}$, $q = 18$ and the green, dashed curve represents a binary of mass $50 M_\odot$, $z = 0.1$ and $q = 1$.

method of Kaiser and McWilliams [97], which we will describe in some detail. The scheme of Peters and Mathews [96] is based on an approximation of Keplerian ellipses. If we assume that the orbital parameters change slowly due to the emission of gravitational radiation, and that the gravitational waves are emitted at the integer multiples of the orbital frequency, the strain amplitude in the n th harmonic can be described as

$$h_n = g(n, e) \frac{G^2 M_1 M_2}{D a c^4} \quad (1)$$

$$\simeq 1.6 \times 10^{-22} g(n, e) \left(\frac{D}{1 \text{ Gpc}} \right)^{-1} \left(\frac{a}{10^{-2} \text{ pc}} \right)^{-1} \left(\frac{M_1}{4 \times 10^4 M_\odot} \right) \times \left(\frac{M_2}{10 M_\odot} \right), \quad (2)$$

where we have normalised to parameters corresponding to an intermediate-mass ratio inspiral (IMRI, see e.g. [29, 31–33, 98]), $g(n, e)$ is a function of the harmonic number n , D being the distance between them, and the eccentricity of the source e . The full frequency evolution of the strain amplitude is discussed in section 4.2. In figure 6 we give an example of an IMRI and the inspiral of a binary of two stellar-mass black holes with masses $30 M_\odot$.

Another interesting example is shown in figure 7. A system of two intermediate-mass black holes (IMBH) and an IMRI which, again, are only observable by LION at these distances and

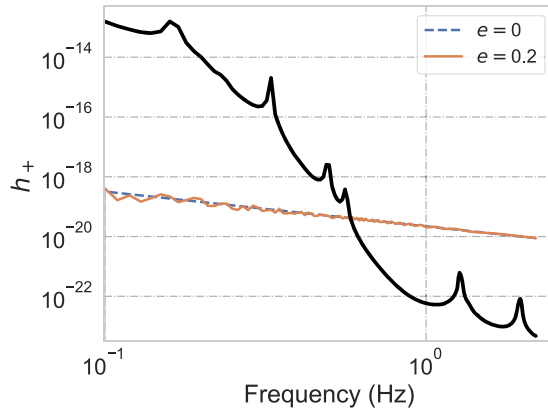


Figure 9. Evolution of the plus polarization h_+ for a binary of two IMBHs of masses $M_1 = M_2 = 10^3 M_\odot$ in the LION window (solid, black line). We consider two different eccentricities, one zero and the other $e = 0.2$ at the beginning of the evolution. As the systems evolve, the loss of angular momentum circularises the eccentric case, which converges toward the circular one. The binaries are assumed to be at a distance of 1 Gpc. For LION, the circular- and eccentric cases yield SNR ~ 1855 and 1875 , respectively, while for (Advanced) LIGO the values are ~ 3.7 for the circular case, and 3.8 for the eccentric one.

parameters. As we can see in the left panel, LION combined with LISA would allow us to do multi-bandwidth gravitational wave astronomy (see e.g. [26]). Since we can observe a source with LISA in the inspiral phase years before the final merger, we can predict the time it is seen in the lion band with an accuracy on the order of a few seconds. This early warning would likely help to enhance the detection of the merger and ringdown with LION. Thanks to this, we would be in the position of breaking various degeneracies in the parameter extraction of the source. As before, we note that only LION is in this unique position.

As noted in [30], there is a need for a deci-Hertz gravitational wave observatory such as LION to understand the formation of compact binaries. This is something that LISA and LIGO/Virgo are not in the position of doing, as explained in [30].

4.2. Realistic waveforms and one comparison to LIGO

After our first exploration of the possible interesting systems for LION to observe, we now address with more realistic waveforms some examples. In particular, we employ the method of [101, 102], commonly referred to as IMRPhenomD, via the approach of [97]. The family IMRPhenomD approaches coalescing binaries by compounding a post-Newtonian approximation for the inspiral with a fully numerical relativistic solution for the merger and ringdown. This hybrid method allows us to quickly compute different waveforms. We also consider the approximation of [103], ENIGMA, which is a time domain, inspiral-merger-ringdown waveform model which allows consideration of moderately eccentric orbits, via the implementation of [104]. In particular, we display in figure 8 four sources, with one of them only in the LISA regime as a reference point.

Probably the most remarkable system is a binary of two black holes of mass ratio $q = 18$ with the heaviest one having a mass of $100 M_\odot$ at a redshift of $z = 100$ (lowermost signal). This binary is only detectable by LION and shows the relevance of the detector.

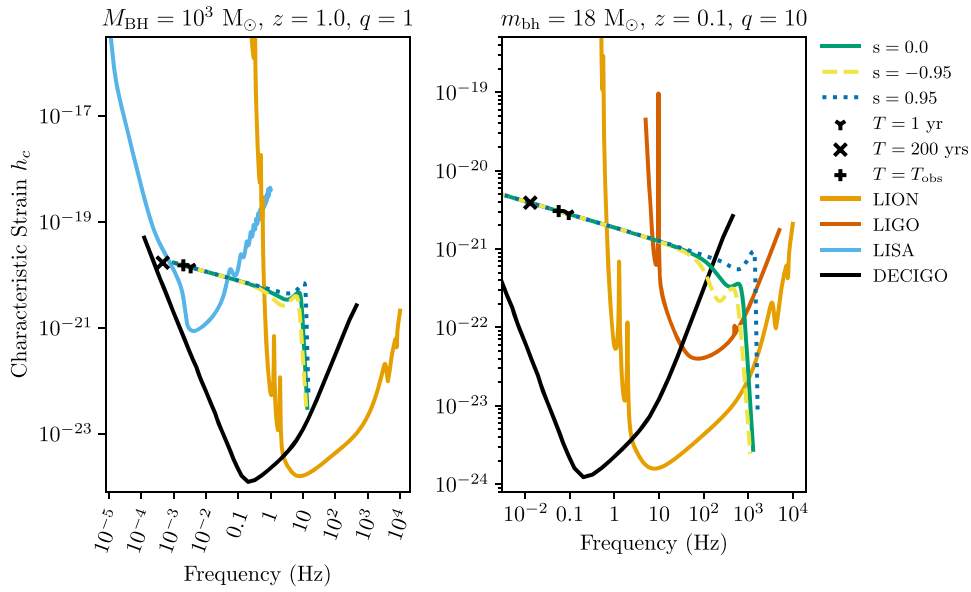


Figure 10. *Left panel:* a binary of two IMBHs of masses $10^3 M_{\odot}$ at a redshift $z = 1$ for different spin values (see legend on the right) at three different moments, namely 200 years before the coalescence, at the detector’s (typical) observation time T_{obs} and 1 year before the coalescence. We display the sensitivity curve of LISA (blue, left one) and LION (orange, right one). *Right panel:* evolution of the waveform for a binary of two black holes with $q = 18$ and the mass of the heaviest one $m_{\text{bh}} = 18 M_{\odot}$, at a redshift $z = 0.1$. As in the left panel, we pinpoint the same moments in the evolution, different spin configurations, the sensitivity curve of LION (orange, left one) and LIGO (red, right one).

We show in figure 9 a binary of two IMBHs with equal mass and different eccentricities, as approximated with ENIGMA. The case with eccentricity yields a slightly higher signal-to-noise ratio (SNR) than that calculated for chirping binaries.

Because IMRPhenomD features mass ratios up to 18 including spins, in figure 10 we show the role of the spin s and q for two different systems, a binary of two IMBHs and a lighter one, with the mass of the heaviest black hole $18 M_{\odot}$, and $q = 10$. As expected, a mass ratio and positive spin value increases the characteristic strain as we approach the end of the evolution, which has an impact on the calculation of the SNR.

4.3. Signal-to-noise ratio contour maps

To estimate the SNR for different configurations and systems, we adopt the approach of [24] and assume that the noise is stationary, Gaussian and uncorrelated with the signal. If we envisage the waveforms as vectors in a Hilbert space [105], we can define a noise-weighted inner product as follows

$$\langle h'|h \rangle := 2 \int_0^{\infty} df \frac{\tilde{h}'(f)\tilde{h}(f)^* + \tilde{h}'(f)^*\tilde{h}(f)}{S_n(f)}. \quad (3)$$

In the last equation we have introduced $\tilde{h}(f)$ as the Fourier transform of the time-domain waveform. Moreover, $S_n(f)$ is the one-sided noise power spectral density of the detector, following

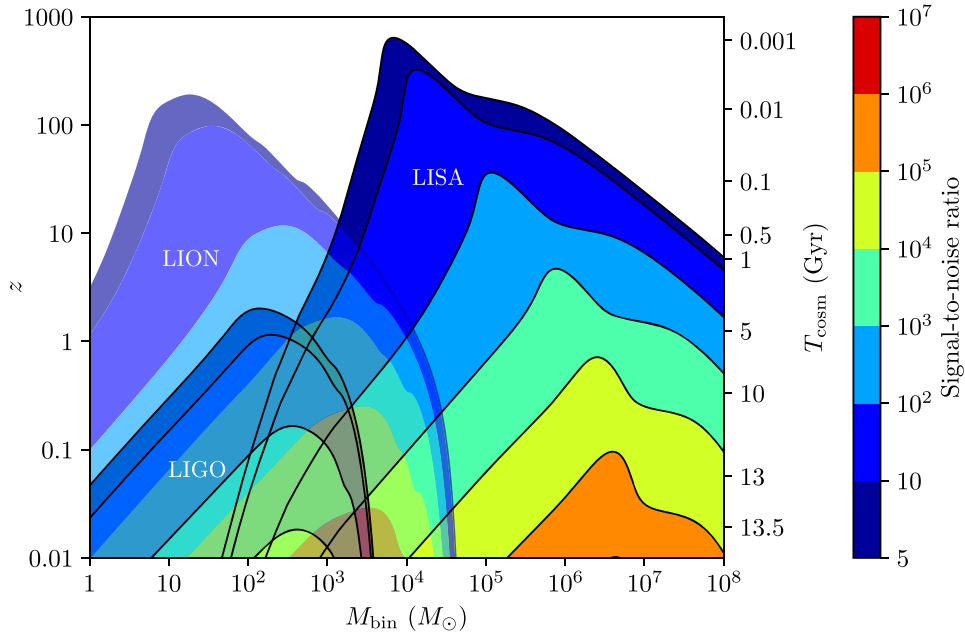


Figure 11. Contours of SNR as a function of the total mass of the binary system M_{bin} in solar masses and redshift (and cosmological time, in Gyrs). We remark how LION covers a region of phase-space which is totally inaccessible to both LISA and LIGO, as shown with particular examples in figure 8.

[106, 107], and see section 2.3 of [97] for a summary and implementation. If we adopt the above-mentioned assumptions, the optimal SNR when filtering the data against h is given by

$$\rho = \sqrt{\langle h_+ | h_+ \rangle + \langle h_\times | h_\times \rangle}. \quad (4)$$

In figure 11 we depict the SNR ρ as a function of the total mass of the binary M_{bin} and redshift for LION, but also LIGO and LISA for a reference point. We note that LION is in the position of extracting information about binaries with masses in the regime necessary to understand the formation and cosmical evolution of supermassive black holes (see, for a review [25]), with SNRs values at redshift exceeding 100.

Finally, in figure 12 we show how the SNR depends on the redshift, mass ratio and spin values for LION and compare it with LIGO as a reference. We note that, contrary to figure 11, the contour map covers SNRs including values from 1.

4.4. Angular resolution

We note that the frequency range of LION allows us to improve the angular resolution of source positioning in the sky, by adding a lunar detector to the LVK collaboration. The angular resolution $\Delta\Omega$ is proportional to the increase of the network area A , $\Delta\Omega \propto 1/A$ [27]. This means that an average Earth–Moon distance of 380 000 km [108] increases the area A and improves the angular resolution by a factor of 30 compared to a purely Earth based network with maximum distance of 12 000 km.

This will not only help to enhance the sky mapping of these objects, but also opens a huge possibility to multi messenger astronomy, as we have seen, in particular in figure 11. The

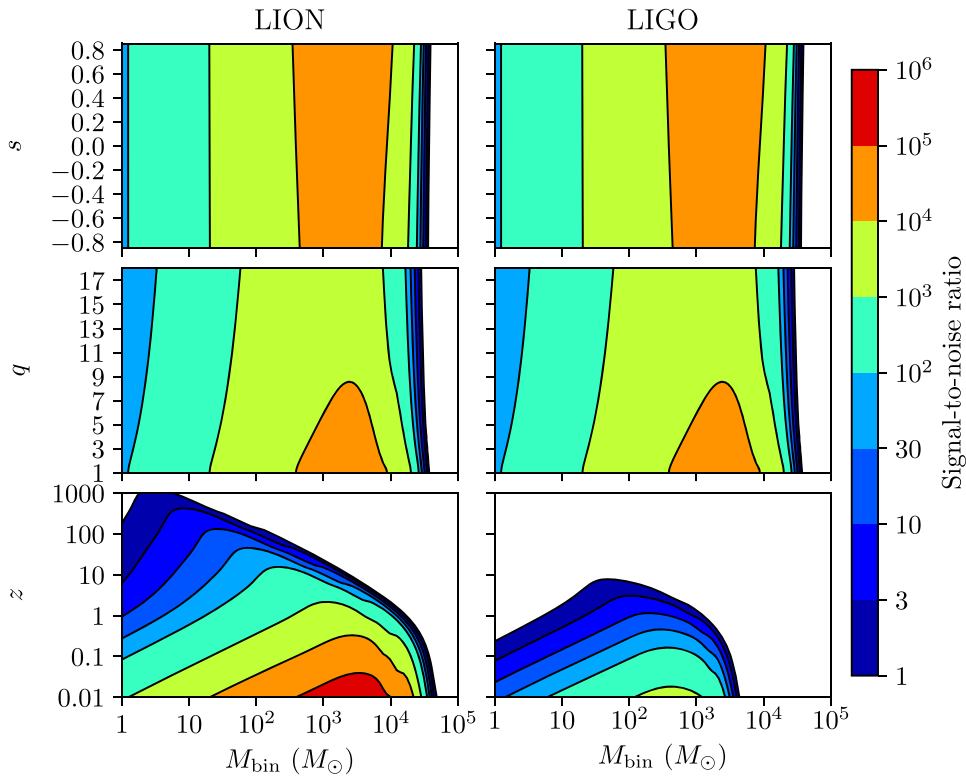


Figure 12. Mosaïque of SNR contour maps to show its dependency with the redshift (bottom panels), as in figure 11), the mass ratio q ranging between the values that IMR-PhenomD allows for (mid panels, see text) and the spin (top panels). We show LION on the left column and LIGO on the right one.

first counterpart observation GW170817 [109], showed, how much knowledge is gained when observing the emitted gravitational waves and electromagnetic radiation from an event. With the combination of improved sky location and knowledge about the time of the merger beforehand, the electromagnetic telescopes can be pointed to the sky location in time to ensure a simultaneous observation. This will significantly increase the upcoming gravitational-wave optical transit observer array’s useful observations [110].

5. Cost and timeline

The costs for LION can only be roughly estimated at this point. We want to stress that LION should focus on using existing gravitational wave detector technologies as well as by then existing space flight technologies. Still, we assume here a conservative budget, highly dependent on technological investigations of the next years. Based upon a study from the Center for Strategic and International Studies [111], an infrastructure on the Moon and its installation, including the launch and landers, can be approximated to cost 30 billion€, from which we exclude the 14 B€ for the Luna base itself. The CE development costs are currently assumed to be about 55 M€ [6]. Assuming, that upgrading vacuum compatibility to space compatibility is a factor of 10 in costs, this leads to 0.55 B€ for the development. Meanwhile, the total costs of LIGO are assumed to be in the order of 500 M€ [112]. Bringing

these numbers together gives us a conservative estimate for LION of $(30 - 14 + 0.55 + 5) \text{Be} = 21.55 \text{Be}$.

With an initial mission proposal by 2023 and a full design study based on new data about the Moon by 2030, simultaneous development of technologies along side CE's second phase would enable us to begin space qualifying the necessary parts so that all technologies are space qualified by 2048. This would give time to build the missions in time for three launches in the middle of the decade. Assembling would be done robotically and it is hoped that this will be completed within a couple of years. The mission lifetime is expected to be 10 years, with the possibility of 5 years expansion. It is expected that in this time components will wear out and limit operation. However, if there proved to be sufficient interest, LION can continuously be repaired and upgraded, making an advanced LION detector more likely than the need of disposal. This, again, is meant to be a conservative estimate. An acceleration of the progress, to overlap with the planned LISA mission launching in 2034 is highly recommended.

6. Conclusion

We find that a detector built on the Moon with technology of the third generation detectors features a detection bandwidth between 0.7 Hz and 10 kHz with detection noise hitting a floor of $3 \times 10^{-24} \text{ m Hz}^{-1/2}$ at 10 Hz. This is a conservative estimate which is limited by the thermal noise and residual seismic activity of the Moon. Since the actual seismic activity of the Moon is still uncertain and we only use an upper estimate of activity, a thorough test campaign is likely to improve the performance prediction. Nevertheless, LION would make an excellent complement to the existing network, with significant overlap of required technology development.

The unique detection band presents an opportunity to push the capabilities of gravitational wave astronomy and our understanding of physics. The ability to measure binaries of seed black holes to a redshifts of 100 and higher provides a unique glance at the cauldron where stars and galaxies were formed and would shed light on the question on how massive black holes assemble their mass and grow over cosmic timescales (see e.g. [25], for a review). We also find that many of the sources present in the LISA band can be picked up in the LION band to observe final merger achieving the long held dream of multi messenger astronomy. Doing so will break many of the degeneracies of the parameter space so that we can better understand their formation channels [30]. In this regard, it can be envisaged as a true deci-Hertz bridge between space borne observatories and ground based ones. Moreover, having a much wider detection array at low frequencies is imperative for the coordination between electromagnetic and gravitational observations needed to glimpse the final stages of an inspiraling system with an optical telescope. Thus, LION will reach its full potential when being operated together with the third generation detectors on the Earth and in space.

The Moon is perhaps one of the best available environments to operate a gravitational wave detector in our Solar System. While we show in this paper that current or currently designed technology enables the low frequency detector, its installation on the lunar surface requires significant technological development. However, these lie not too far beyond the horizon of humanities progress. We believe that, once humanity starts exploiting the Moon for science and resources, a gravitational wave observatory is an excellent candidate for the first large scale infrastructure experiment on the lunar surface.

Author contributions

The initial idea of LION was developed by the authors Bischof, Carter, Hartig and Wilken at a lecture week in 2018, where Amaro-Seoane was supervising, run by the International Max Planck Research School for Gravitational Waves (IMPRS-GW). Bischof, Carter, Hartig and Wilken followed up the lecture week with a deeper exploration of the required mission parameters, noise curves and technical discussion of the paper. The work on sources in the detection band was performed by Pau Amaro Seoane.

Acknowledgments

This paper builds on a student project from an lecture week by the International Max-Planck Research School for Gravitational Waves (IMPRS-GW) in 2018. Therefore we would like to thank the IMPRS-GW for organisation and Karsten Danzmann for providing the space necessary for the development of this research. We thank Stefan Danilishin and James Lough for the discussion of the technical and the environmental challenges. We are also thankful to Christoph Vorndamme for discussion on the initial noise budgets. We thank Frank Ohme for his comments and suggestions, as well as Andrew Kaiser for comments and for making his approach publicly available. Finally, we would like to thank Harald Lück and Jens Reiche for useful feedback on the manuscript. We also acknowledge that the work of the authors Bischof, Carter, Hartig and Wilken was supported by the IMPRS-GW grant and the funding by the Deutsche Forschungsgemeinschaft in the Collaborative Research Center SFB1128 Relativistic Geodesy and Gravimetry with Quantum Sensors (geo-Q) at Leibniz Universität Hannover. Bischof, Carter, Hartig and Wilken gratefully acknowledge support by the Deutsches Zentrum für Luft-und Raumfahrt (DLR) with funding from the Bundesministerium für Wirtschaft und Technologie (FKZ 500Q1801). Amaro-Seoane acknowledges support from the Ramón y Cajal Program of the Ministry of Economy, Industry and Competitiveness of Spain, as well as the financial support of Programa Estatal de Generación de Conocimiento (reference PGC2018-096663-B-C43) (MCIU/FEDER), and the COST Action GWverse CA16104. His work was supported by the National Key R & D Program of China (2016YFA0400702) and the National Science Foundation of China (11721303).

Data availability statement

The data that support the findings of this study are available upon reasonable request from the authors.

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