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Sail Performance Analysis of Sailing Yachts by Numerical Calculations and Experiments

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1. Introduction

Sails of a sailing yacht can be considered as multiple soft thin wings (membrane wings) with relative large cambers, and are often used at large attack angles. The shape of sail is determined as an equilibrium state of both aerodynamic force and tension acting on the sail surface. In particular, a spinnaker used for the running condition is a very soft membrane like a parachute, and the shape is simply formed by self-generated aerodynamic forces which are strongly affected by the sail shape itself. These facts lead to new challenges in the present problem, i.e., in the measurements the sail shape must be accurately measured in the flying condition, and in numerical simulation of flow and forces the sail flying shape is correctly given or predicted as a part of solution. The present study concerns the authors' ongoing effort on analyses of sail performance of sailing yachts by numerical calculations and experiments, and in this paper, the focus of discussions is more on the former. Two computational fluid dynamics (CFD) methods are used in the present study, and the results are validated through detailed comparison with experimental data. The data are obtained in onboard full-scale measurements by using a sail dynamometer boat. Our study concerns both the upwind and downwind sailing conditions; however, we focus on the former in the present chapter due to the limitation of space in this book. More detailed background of the present work is well described in Masuyama et al. (2009).

One of the two CFD methods is a Vortex Lattice method (VLM). Although the VLM is a potential flow calculation, it is well known the results agree well with the measured data at the upwind condition of small attack angle. The VLM is used as the sail design and making tool due to the quick convergence ability for the parametric survey of sail shape to obtain the desired sail performance, and also due to good compatibility with the finite element method (FEM) for the strength analysis. In this paper, a method to shed wake vortices step-by-step developed by Fukasawa was adopted in the Vortex Lattice method (Fukasawa, 1993; Fukasawa & Katori, 1993).

Another CFD method is a Multiblock Reynolds-Averaged Navier-Stokes (RANS)-based CFD named "FLOWPACK". This code was developed by Tahara specifically for CFD education and research, and design applications for ship hydrodynamics, aerodynamics, and fluid engineering (Tahara, 2008). As part of the developments for application to design problems, a complete multiblock domain decomposition feature was included. The numerical method of FLOWPACK solves the unsteady RANS and continuity equations for mean velocity and pressure. Either a zero or a two-equation turbulence model can be used for turbulence flow calculation, and in the present study the former was used. The FLOWPACK was included as a

solver in a sail performance analyzer named “*Advanced Aero Flow (AAF)*” developed by Katori (Katori, 2009). The AAF is a specialized package for the calculation of sail performance of sailing yachts, and composed of both mesh generator and post analyzer.

The sail shapes and performance were measured using a sail dynamometer boat *Fujin* under sailing condition on the sea (Masuyama et al., 1997a, 1997b). *Fujin* is a 34-foot LOA boat, in which load cells and CCD cameras were installed to simultaneously measure the sail forces and shapes. At the same time, the sailing conditions of the boat, e.g., boat speed, heel angle, wind speed, and wind angle, were measured. The shapes and 3D coordinates of the sails were used for the input data of the numerical calculations, and the calculated results were compared with the measured data. The sail coordinates with aerodynamic coefficients are tabulated for some sailing conditions in order to provide benchmark data for the CFD validation.

In this paper, overview of the above-mentioned CFD methods and experiments are described. As the aforementioned, sail flying shapes are considered in the present CFD so that the accurate prediction of flow and aerodynamic forces is possible. Discussion of the results is based on detailed comparison with the measurements. The discussion also includes the current capability of the CFD methods in the present problem, and prognosis for the enhancement of the capability in future work for higher accuracy and/or more complicated flow simulation. It will be noteworthy that the overall trends of the flow and the aerodynamic forces measured in the experiments are fairly well predicted by the present computations; and at the same time, experimental techniques originally implemented and used in the present study are shown very promising and capable to provide very detailed benchmark data for CFD validation.

2. Sail plan for the analysis

In this study the experiments and numerical calculations were performed for the upwind sailing condition. The sail shapes and performance were measured using a sail dynamometer boat *Fujin*. The sail plan of the *Fujin* and the coordinate system are shown in Fig.1. The principal dimensions of the boat and the detailed measurements of the sails are also shown in Table 1. The measurement system of the boat and testing conditions are described in section 5, and the measured and calculated results are compared and discussed in section 6.

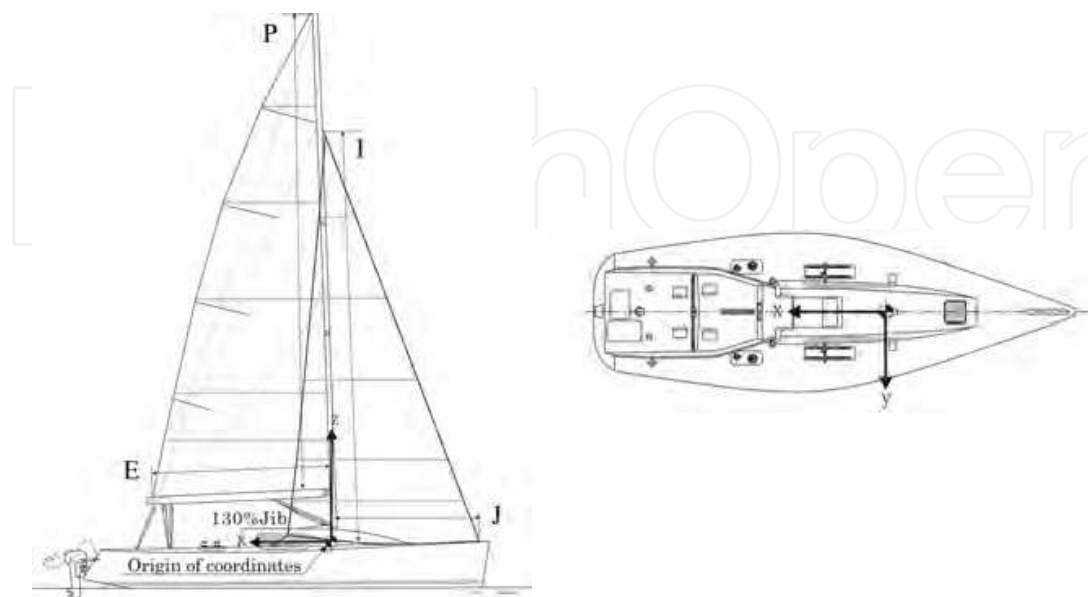


Fig. 1. Schematic showing the sail plan of *Fujin* with 130% jib and the coordinate system

HULL		SAIL DIMENSIONS		
Length Over All [m]	10.35		Mainsail	130% Jib
Length Water Line [m]	8.80	Peak Height [m]	13.82	10.70
Breadth Maximum [m]	3.37	Luff Length [m]	12.50	11.45
Breadth Water Line [m]	2.64	Foot Length [m]	4.44	4.89
Displacement [ton]	3.86	Sail Area [m ²]	33.20	26.10
SAIL		Height [%]	Chord Length [m]	
I [m]	11.00	0	4.44	4.89
J [m]	3.61	10	4.13	4.44
P [m]	12.55	20	3.85	3.94
E [m]	4.51	40	3.23	2.94
I, J, P, E of Sail are defined in Fig. 1.		60	2.43	1.97
		80	1.39	0.98
		100	0.15	0.10

Table 1. Principal dimensions of *Fujin* and detailed measurements of sails

3. Overview of Vortex Lattice method (VLM)

3.1 Basic concept of Vortex Lattice method

The Vortex Lattice method is a branch of CFD, and it is often used at the early stage of yacht sail design because of the comparatively less computational time. This method is based on the potential theory, similar to the panel method, and the flow around the sail is expressed by discrete vortices. The Vortex Lattice method has its root in the lifting line theory formulated by Prandtl in 1918. A wing is represented by a single vortex line in the lifting line theory, and the force acting on the wing is approximated by the force acting on the vortex line.

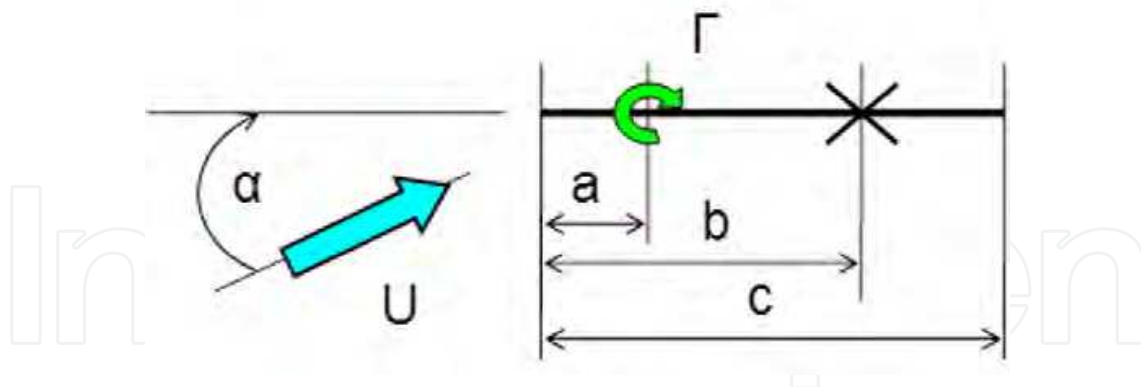


Fig. 2. 2-Dimensional flat plate wing and a vortex filament

Firstly, 2-dimensional flow around a flat plate wing is considered. A vortex filament is located at a distance "a" from the leading edge of the wing as shown in Fig.2. Although the onset flow U is constant, the flow over the wing is accelerated, while it is decelerated below the wing, because of the flow induced by the vortex filament. This leads to the pressure decrease on the back surface and the pressure increase on the front surface of the wing accordingly to Bernoulli's theorem. This means that the flow around the wing can be realized by a vortex filament in the flow. The strength of vortex filament, or sometimes called circulation, is determined by a boundary condition on the wing; that is, there is no cross flow through the

wing. This boundary condition is usually satisfied at a certain point called control point. Assuming that the control point is located at a distance " b " from the leading edge of the wing shown in Fig.2, and satisfying the boundary condition at this point, the strength of the vortex filament can be determined. Once the strength of vortex filament is determined, the lift acting on the vortex filament can be calculated according to Kutta-Joukowski theorem, that is,

$$L = \rho U \Gamma \quad (1)$$

where ρ is the density of the fluid and Γ is the strength of vortex filament. If the calculated lift is assumed to equal that generated in a 2-dimensional thin parabolic shape airfoil, the locations of the vortex filament and the control point are determined to be $a = c/4$ and $b = 3c/4$, where " c " is the chord length of the wing. This is called 1/4-3/4 rule, which was shown by Pistoletti (Pistoletti, 1937). This rule is used as the basis of the present Vortex Lattice method.

3.2 Application to sail configuration

As the yacht sail is a 3-dimension shape body, attention should be paid to the treatment of the end of vortex line. According to the Helmholtz's theorem on vortex, the vortex line should expand from the boundary to the boundary of the flow or shuts oneself and makes vortex ring. Accordingly, in the 3-dimensional body, the vortex line should expand infinity from the edge of the body. In the lifting line theory, or the Vortex Lattice method, the vortex line is assumed to be a horseshoe type shown in Fig.3(a), and the vortex line changes its direction at the edge of the body to extend to infinity as trailing free vortices.

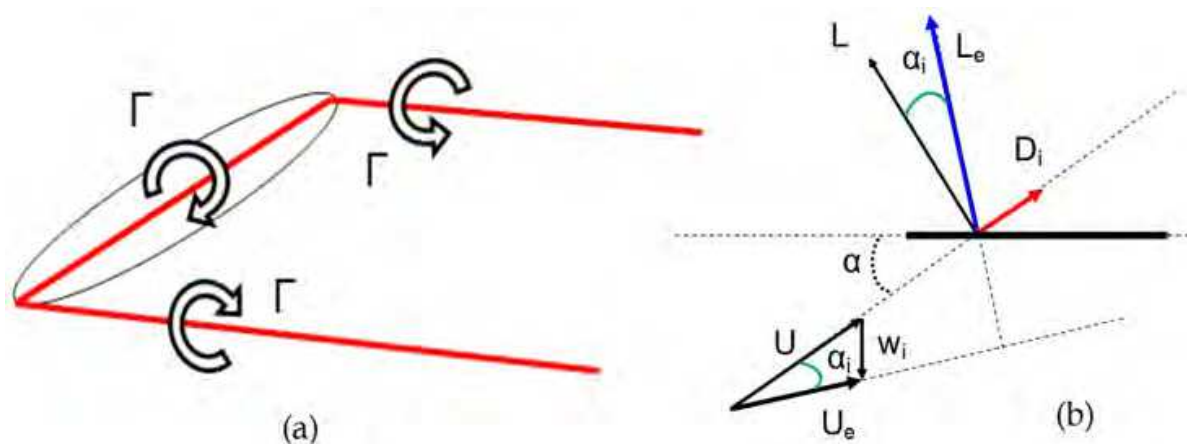


Fig. 3. Horseshoe vortex and downwash effect

In case of the horseshoe vortex shown in Fig.3(a), the flow induced by each vortex line affects the onset flow in the magnitude and the direction. According to Biot-Savart law, the velocity vector induced by a slight part of the vortex line $d\ell$ is given by

$$\vec{v} = \frac{\Gamma}{4\pi} \oint \frac{d\vec{\ell} \times \vec{r}}{|\vec{r}|^3} \quad (2)$$

where \vec{r} is a position vector from the vortex part to the point concerned. The downward velocity called downwash w_i is calculated by using Equation (2), which affects the onset flow. This causes the reduction of the attack angle of total inflow by α_i as shown in Fig.3(b),

and the total velocity of inflow into the wing changes to U_e . If the lift acting on the wing is defined as the force perpendicular to the onset flow direction, it is given by the following formula according to Kutta-Joukowski theorem.

$$L = L_e \cos \alpha_i \approx L_e = \rho U_e \Gamma \approx \rho U \Gamma \quad (3)$$

In this case, the force in the onset flow direction is generated, which is the apparent drag called induced drag given by

$$D_i = L_e \sin \alpha_i \approx L_e \alpha_i = \frac{L_e}{U} w_i \approx \frac{L}{U} w_i \quad (4)$$

The induced drag is a distinctive drag in a 3-dimensional wing, and does not appear in a 2-dimensional wing.

In the Vortex Lattice method, the lift, induced drag, and center of pressure are calculated by arranging horseshoe vortices of different strength on the surface of sail. By placing a number of horseshoe vortices, the sail of complex shape with twist, camber, or two or more sails, can be analyzed. Falkner used the name "Vortex Lattice" firstly in his report, in which a wing was covered with a grid of straight horseshoe vortices (Falkner, 1943, 1946). In 1950's, only the analysis where the trailing vortices are placed in the straight line was able to be carried out because of the computer capability, and the accuracy was questionable. It was 1965 when the Vortex Lattice method started to demonstrate its ability along with the development of computer, and the method came to be used for the performance prediction of yacht sail. The yacht sail is one of the most suitable objects for applying the Vortex Lattice method because of its thickness, if the viscous effect of fluid can be disregarded. An application of the Vortex Lattice method to the performance prediction of yacht sail will be explained in the following paragraph with the use of a step-by-step procedure to estimate the trailing vortex deformations.

Discretized horseshoe vortices are located on the sail plane in the Vortex Lattice method. It is usual to divide the sail plane into quadrilateral panels as shown in Fig.4, and the

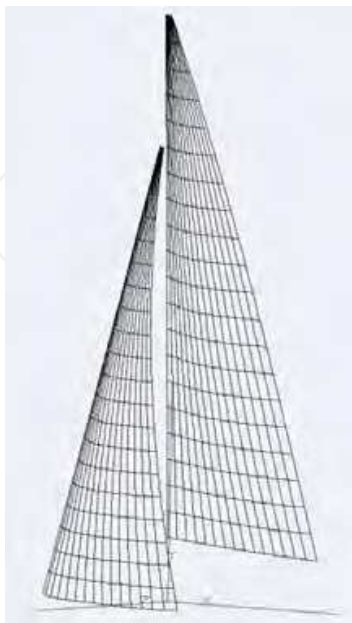


Fig. 4. Panel discretization of sails

horseshoe vortices are placed at 1/4 length of panel from the front end edge of the panel so as to trail the trailing vortices rearwards. The strengths of the horseshoe vortices are determined by satisfying the boundary condition on the sail; that is, the total flow of the onset flow and the induced wake by vortices is parallel to the sail surface at control points. The control point is taken to be the point 3/4 of length of panel from the front end edge of the panel according to the 1/4-3/4 rule.

According to Biot-Savart law, the velocity vector at the control point of i -th panel induced by other vortices are given by

$$\vec{v}_i = \frac{1}{4\pi} \int \frac{\Gamma (\vec{d}\vec{\ell} \times \vec{r})}{|\vec{r}|^3} = \sum_j \frac{\Gamma_j}{4\pi} \sum_{k=1}^3 \frac{\cos\alpha_{kji} + \cos\beta_{kji}}{h_{kji}} \vec{e}_{kji} \quad (5)$$

where α_{kji} , β_{kji} , h_{kji} , are the angles and the distance of a k -th filament of a j -th horseshoe vortex and i -th control point shown in Fig.5, which shows a plane containing the vortex filament and the control point. \vec{e}_{kji} is a unit vector perpendicular to the plane shown in Fig.5. $k=1, 2, 3$ in Equation (5) denotes each vortex filament of a horseshoe vortex. With the use of Equation (5), the boundary condition on the control point can be given by

$$\sum_i^{NB} \vec{v}_i \cdot \vec{n} = \vec{U} \cdot \vec{n}_i - \sum_i^{NW} \vec{v}_i \cdot \vec{n} \quad (6)$$

where \vec{U} is an onset flow velocity vector and \vec{n} is the unit normal vector at the control points. NB is the number of bound horseshoe vortices on the sail plane and NW is the number of trailing horseshoe vortices in the wake. Equation (6) can be written in the vector matrix form by

$$[\Lambda]\{\Gamma\} = \{\mathbf{u}_n\} + \{\mathbf{v}_n\} \quad (7)$$

Solving Equation (7), the strength of bound vortices can be obtained.

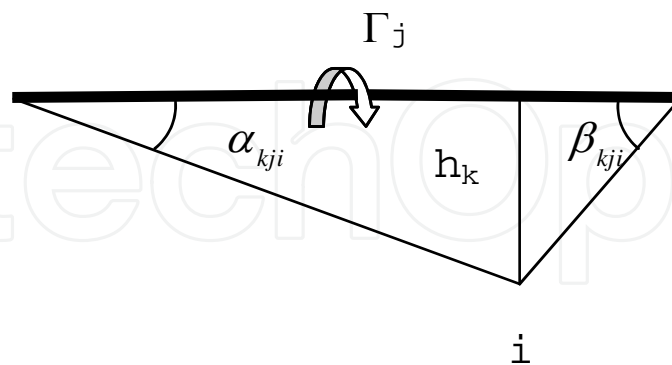


Fig. 5. K -th filament of j -th horseshoe vortex and i -th control point

3.3 Step-by-step vortex shedding technique

The important point in the Vortex Lattice approach to yacht sail is the handling of the wake of the sail. The wake vortices proceed downstream from the trailing edge, or leech/foot of sail, in the Vortex Lattice method. The location of wake vortices are determined by the condition that they are free vortices; that is, the stream line of wake vortices should be

parallel to the velocity field induced by total vortex system. A step-by-step procedure developed by Fukasawa was adopted in this paper to determine the strength of the bound vortices and the location of wake vortices (Fukasawa, 1993; Fukasawa & Katori, 1993; Masuyama et al., 1997a, 1997b). The wake vortices are shed from the trailing edge in each time step according to Helmholtz's theorem; that is,

$$\Gamma_B + \Gamma_W = 0 \quad (8)$$

$$\frac{D\Gamma_B}{Dt} = 0 \quad (9)$$

where Γ_B and Γ_W are the total strength of bound vortices, or the circulation around the sail, and the strength of wake vortices, respectively. From Equation (9), we have

$$\frac{\partial \Gamma_B}{\partial t} = \tilde{U} \frac{\partial \Gamma_B}{\partial \xi} \quad (10)$$

and substituting Equation (10) into (8), we have

$$\frac{\partial \Gamma_B}{\partial t} = -\tilde{U} \frac{\partial \Gamma_W}{\partial \xi} \quad (11)$$

where ξ is taken to the downstream direction, and \tilde{U} is the local velocity at the wave vortex. Assuming that the wake vortices proceeds $\Delta\xi$ downstream in a time step Δt with the velocity \tilde{U} , the strength of wake vortex shed at time step k , can be given by integrating Equation (11), that is,

$$\Delta\Gamma_W^k = -\frac{1}{\tilde{U}} \int_0^{\Delta\xi} \frac{\partial \Gamma_B}{\partial t} d\xi = -\frac{\Delta\xi}{\tilde{U}\Delta t} (\Gamma_B^k - \Gamma_B^{k-1}) = \Gamma_B^{k-1} - \Gamma_B^k \quad (12)$$

Equation (12) means that the strength of wake vortex shed at time step k is the increase of the strength of bound vortex from time step $k-1$ to time step k . Once a vortex filament is shed at time step k , it proceeds downstream with a constant strength according to the local field velocity, i.e., each horseshoe wake vortex moves in the direction of field velocity in each time step. The field velocity is updated in every time step. The calculation is carried forward until the the calculated lift and drag forces converges. The forces vector and the moment acting on the sail are calculated accordingly to Kutta-Joukowski theorem.

$$\vec{F} = \rho \int \Gamma \tilde{U} \times d\vec{s} \quad (13)$$

with the use of the vortex strengths of the wake vortices and the bound vortices determined by solving Equation (7).

Finally, the overall numerical solution procedure of the present Vortex Lattice method is summarized as follows:

- Step 1. Divide the sail planes into quadrilateral panels, and allocate horseshoe vortices on the sail plane.
- Step 2. Input the mast rake angle, heel angle of the yacht, apparent wind speed and apparent wind angle.

- Step 3. Solve the strength of bound horseshoe vortices on the sail plane with non-deformed trailing vortices.
- Step 4. Compute the total circulation around the sail caused by horseshoe vortices.
- Step 5. Compute the increment of the total circulation and shed free horseshoe vortices according to Equation (12).
- Step 6. Calculate the local velocities along the wake, and deform the trailing vortices.
- Step 7. Compute the force vector and moment acting on the sails.
- Step 8. Solve the strength of bound horseshoe vortices on the sail plane with trailing free vortices.
- Step 9. Repeat Step 4 through Step 8 until the force is converged.

Fig.6 shows the example calculation results. In the present study, the mast and rigging were not considered for the series calculations, and the mirror image was taken into account about the deck plane of the boat. Since the vortex lattice methods do not predict viscous drag, the viscous drag acting on the sails and rigging was calculated empirically using a drag coefficient C_{Dp} . The value of C_{Dp} was obtained from the measured data in the previous papers and formulated for the upwind condition as follows:

$$C_{Dp} = 0.0026 \gamma_A + 0.005 \quad (14)$$

where γ_A is apparent wind angle in degrees.

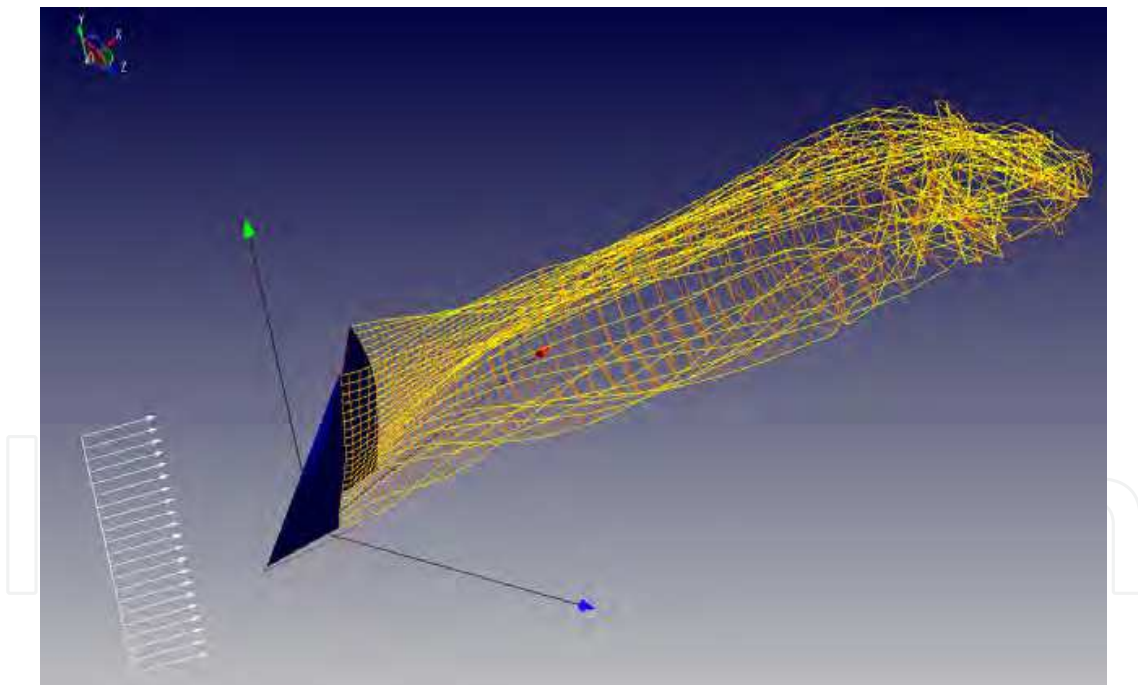


Fig. 6. Calculated wake by Vortex Lattice method

4. Overview of Reynolds-averaged Navier-Stokes equation method

The RANS-based CFD method used in the present study was *FLOWPACK*. The code was developed by Tahara specifically for CFD education and research and for design applications for ship hydrodynamics, aerodynamics, and fluid engineering. As part of the developments for application to design problems, a complete multiblock domain

decomposition feature was included. At present, *FLOWPACK* has a good interface with the authors' inhouse automatic grid generator as well as with commercial grid generation software. For a complete documentation of the method is available in Tahara (2008). In the following, an overview of the numerical method is given.

4.1 Governing equations

Let us consider a sail system fixed in the uniform onset flow (see Fig.1 for the basic coordinate system). The non-dimensional RANS equations for unsteady, three-dimensional incompressible flow can be written in Cartesian tensor notation as

$$\frac{\partial U_i}{\partial t} + U_j \frac{\partial U_i}{\partial x^j} + \overline{\frac{\partial u_i u_j}{\partial x^j}} + \frac{\partial p}{\partial x^i} - \frac{1}{\text{Re}} \nabla^2 U_i = 0 \quad (15)$$

$$\frac{\partial U_i}{\partial x^i} = 0 \quad (16)$$

where U_i ($i=1,2,3$) = (U, V, W) and u_i ($i=1,2,3$) = (u, v, w) are the Cartesian components of mean and fluctuating velocities, respectively, normalized by the reference velocity U_0 , x^i ($i=1,2,3$) = (X, Y, Z) is the dimensionless coordinates normalized by a characteristic length L , $\text{Re} = U_0 L / \nu$ is the Reynolds number, ν is the kinematic viscosity, the barred quantities $\overline{u_i u_j}$ are the Reynolds stresses normalized by U_0^2 , and p is the pressure normalized by ρU_0^2 . If $\overline{u_i u_j}$ are related to the corresponding mean rate of strain through an isotropic eddy viscosity ν_t , i.e.,

$$\overline{u_i u_j} = \nu_t \left(\frac{\partial U_i}{\partial x^j} + \frac{\partial U_j}{\partial x^i} \right) - \frac{2}{3} \delta_{ij} k \quad (17)$$

where $k = (\overline{uu} + \overline{vv} + \overline{ww}) / 2$ is the turbulent kinetic energy, Equation (15) becomes

$$\frac{\partial U_i}{\partial t} + \left(U_j - \frac{\partial \nu_t}{\partial x^j} \right) \frac{\partial U_i}{\partial x^j} - \frac{\partial \nu_t}{\partial x^j} \frac{\partial U_j}{\partial x^i} + \frac{\partial p}{\partial x^i} \left(p + \frac{2}{3} k \right) - \frac{1}{R_\phi} \nabla^2 U_i = 0 \quad (18)$$

where $1/R_\phi = 1/\text{Re} + \nu_t$, and $\phi = U_i$ ($i=1,2,3$). Equations (16) and (18) can be solved for U_i and p when a suitable turbulence model is employed to calculate the eddy-viscosity distribution. Either a zero or a two-equation turbulence model can be used for turbulent flow calculation, and a model used for the present study is the former, i.e., Baldwin-Lomax model (Baldwin & Lomax, 1978), which is an algebraic scheme that makes use of a two-layer isotropic eddy-viscosity formulation. Detailed validation study of this model for boundary layer flows around three-dimensional bodies was done by the author (Tahara, 1995; Tahara & Stern, 1996). In this model, the eddy viscosity is evaluated as follows:

$$\begin{cases} (\nu_t)_{inner} & y \leq y_c \\ (\nu_t)_{outer} & y > y_c \end{cases} \quad (19)$$

where y is the distance normal to the wall surface and y_c is the minimum value of y where both the inner and outer viscosities match. The inner viscosity follows the Prandtl-Van

Driest formula, i.e., $(\nu_t)_{\text{inner}} = \lambda^2 |\omega|$, where $\lambda = \kappa y [1 - \exp(-y^+/A^+)]$ is the turbulent length scale for the inner region, κ and A^+ are model constants, $|\omega|$ is the vorticity magnitude, and y^+ is the dimensionless distance to the wall. In the outer region, eddy viscosity is given by $(\nu_t)_{\text{outer}} = K C_{\text{cp}} F_{\text{wake}} F_{\text{Kleb}}$, where K and C_{cp} are model constants, $F_{\text{wake}} = \min(y_{\text{max}} F_{\text{max}}, C_{\text{wk}} y_{\text{max}} U_{\text{dif}}^2 / F_{\text{max}})$, and $F_{\text{Kleb}} = [1 + 5.5(C_{\text{Kleb}} y / y_{\text{max}})^6]^{-1}$. The F_{max} and y_{max} are determined by the value and corresponding location, respectively, of the maximum of $F = y |\omega| [1 - \exp(-y^+/A^+)]$. The quantity U_{dif} is the difference between maximum and minimum velocity magnitudes in the profile and is expressed as $U_{\text{dif}} = (U^2 + V^2 + W^2)_{\text{max}}^{1/2} - (U^2 + V^2 + W^2)_{\text{min}}^{1/2}$. C_{Kleb} and C_{wk} are additional model constants. Numerical values for the model constants are $A^+ = 26$, $\kappa = 0.4$, $K = 0.0168$, $C_{\text{cp}} = 1.6$, $C_{\text{wk}} = 1.0$, and $C_{\text{Kleb}} = 0.3$.

4.2 Discretization and velocity-pressure coupling

In the following, discretization and velocity-pressure coupling of the present RANS method are described. First, it is convenient to rewrite the transport equations for momentum (U_i) in the following general form:

$$\nabla^2 \phi = R_\phi \left[\sum_{j=1}^3 \left(U_j - \frac{1}{\sigma_\phi} \frac{\partial v_t}{\partial x^j} \right) \frac{\partial \phi}{\partial x^j} + \frac{\partial \phi}{\partial t} \right] + s_\phi \quad (20)$$

where ϕ again represents any one of the convective transport quantities (U_i), and s_ϕ is the source function for the corresponding quantity. We transform the physical space (x^i, t) into a rectangular region in the computational space (ξ^i, τ) using the following coordinate transformations:

$$t = \tau, \quad x^i = x^i(\xi^j), \quad e_i \cdot \nabla \phi = \frac{1}{J} \sum_{j=1}^3 b_i^j \frac{\partial \phi}{\partial \xi^j}$$

$$\nabla^2 \phi = \sum_{i=1}^3 \sum_{j=1}^3 g^{ij} \frac{\partial^2 \phi}{\partial \xi^i \partial \xi^j} + \sum_{j=1}^3 f^j \frac{\partial \phi}{\partial \xi^j}, \quad \frac{\partial \phi}{\partial t} = \frac{\partial \phi}{\partial \tau} - \frac{1}{J} \sum_{i=1}^3 \sum_{j=1}^3 b_i^j \frac{\partial x^i}{\partial \tau} \frac{\partial \phi}{\partial \xi^j}$$

Then the continuity equation (16) and the transport equations (20) for momentum parameters can be written as

$$\frac{1}{J} \sum_{i=1}^3 \sum_{j=1}^3 \frac{\partial}{\partial \xi^i} (b_j^i U_j) = 0 \quad (21)$$

$$\sum_{j=1}^3 \left(g^{jj} \frac{\partial^2 \phi}{\partial \xi^j \partial \xi^j} - 2a_\phi^j \frac{\partial \phi}{\partial \xi^j} \right) = R_\phi \frac{\partial \phi}{\partial \tau} + S_\phi \quad (22)$$

where

$$2a_\phi^j = \frac{R_\phi}{J} \sum_{n=1}^3 b_n^j \left(U_n - \frac{\partial x^n}{\partial \tau} - \frac{1}{J \sigma_\phi} \sum_{m=1}^3 b_n^m \frac{\partial v_t}{\partial \xi^m} \right) - f^j$$

$$S_\phi = s_\phi - 2 \left(g^{12} \frac{\partial^2 \phi}{\partial \xi^1 \partial \xi^2} + g^{13} \frac{\partial^2 \phi}{\partial \xi^1 \partial \xi^3} + g^{23} \frac{\partial^2 \phi}{\partial \xi^2 \partial \xi^3} \right)$$

The geometric coefficients b_i^j , g^{ij} , and f^j appearing in the above equations are defined by Thompson et al. (1985). The transport equations (22) on a computational cell (shown in Fig.7(a)) are linearized and evaluating coefficients and source term at the center node P of the element yields

$$\sum_{j=1}^3 \left(g_P^{jj} \frac{\partial^2 \phi}{\partial \xi^j \partial \xi^j} - 2(a_\phi^j)_P \frac{\partial \phi}{\partial \xi^j} \right) = (R_\phi)_P \frac{\partial \phi}{\partial \tau} + (S_\phi)_P \quad (23)$$

or

$$g_P^{11} \phi_{\xi^1 \xi^1} + g_P^{22} \phi_{\xi^2 \xi^2} + g_P^{33} \phi_{\xi^3 \xi^3} = 2(C_\phi)_P \phi_{\xi^1} + 2(B_\phi)_P \phi_{\xi^2} + 2(A_\phi)_P \phi_{\xi^3} + (R_\phi)_P \phi_t + (S_\phi)_P \quad (24)$$

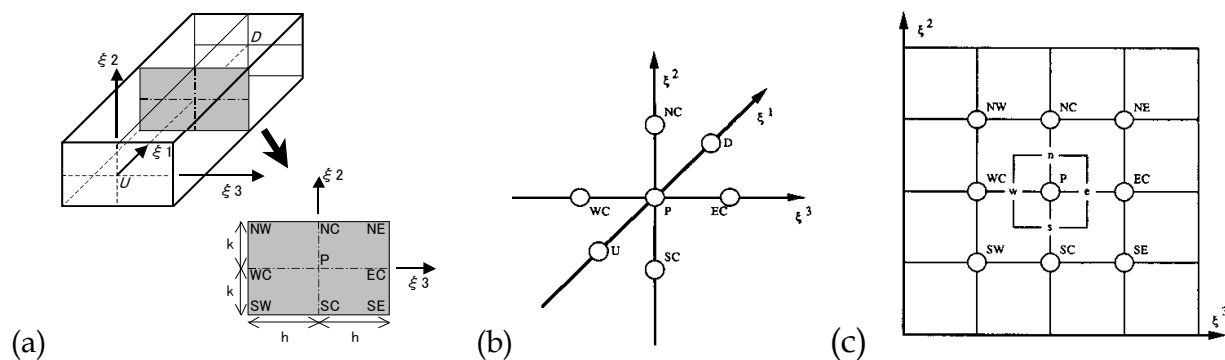


Fig. 7. Definition sketch of a computational cell (a), and nodes in regular grid (b) and continuity cell (c).

The dimensions of the computational cell are $2l \times 2k \times 2h$, where $l=1/\sqrt{g_P^{11}}$, $k=1/\sqrt{g_P^{22}}$, and $h=1/\sqrt{g_P^{33}}$. The above equation is discretized by the finite-analytic scheme. Solution dependent coefficients are analytically derived by solving the above linearized transport equation using a hybrid method which combines a two-dimensional analytic solution in $\xi^2 \xi^3$ -plane with one dimensional analytic solution in the ξ^1 direction. By specifying boundary conditions on the faces of the cell as a combination of exponential and linear functions, which are the natural solutions for the linearized transport equation, Equation (24) can be solved by the method of separation of variables. When the solution is evaluated at the center node P of the element, the following twelve-point finite analytic formula is obtained:

$$\phi_P^n = \frac{1}{1 + C_P \left(C_U + C_D + \frac{R_\phi}{\Delta \tau} \right)} \left[\begin{array}{l} C_{NE} \phi_{NE}^n + C_{NW} \phi_{NW}^n + C_{SE} \phi_{SE}^n + C_{SW} \phi_{SW}^n \\ + C_{EC} \phi_{EC}^n + C_{WC} \phi_{WC}^n + C_{NC} \phi_{NC}^n + C_{SC} \phi_{SC}^n \\ + C_P \left(C_U \phi_U^n + C_D \phi_D^n + \frac{R_\phi}{\Delta \tau} \phi_P^{n-1} \right) - C_P (S_\phi)_P \end{array} \right] \quad (25)$$

where

$$\begin{aligned}
C_U &= \frac{Ce^{Cl}}{l \sinh Cl}, \quad C_D = \frac{Ce^{-Cl}}{l \sinh Cl}, \quad C_{SC} = \frac{e^{Bk}}{2 \cosh Bk} P_A, \quad C_{NC} = e^{-2Bk} C_{SC} \\
C_{WC} &= \frac{e^{Ah}}{2 \cosh Ah} P_B, \quad C_{EC} = e^{-2Ah} C_{WC}, \quad C_{SW} = \frac{e^{Ah+Bk}}{4 \cosh Ah \cosh Bk} (1 - P_A - P_B) \\
C_{SE} &= e^{-2Ah} C_{SW}, \quad C_{NW} = e^{-2Bk} C_{SW}, \quad C_{NE} = e^{-2Ah-2Bk} C_{SW} \\
C_P &= \frac{h \tanh Ah}{2A} (1 - P_A) = \frac{k \tanh Bk}{2B} (1 - P_B), \quad P_A = 4E_2 Ah \cosh Ah \cosh Bk \coth Ah \\
P_B &= 1 + \frac{Bh \coth Bk}{Ak \coth Ah} (P_A - 1), \quad E_2 = \sum_{m=1}^{\infty} \frac{-(-1)^m (\lambda_m h)}{[(Ah)^2 + (\lambda_m h)^2]^2 \cos \sqrt{A^2 + B^2 + \lambda_m^2} k} \\
\lambda_m h &= \left(m - \frac{1}{2}\right) \pi
\end{aligned}$$

The subscripts P, U and D denote the center, upstream and downstream nodes, respectively, and NC, NW, WC, etc. denote the nodes in the $\xi^2 \xi^3$ -plane in terms of compass directions. The superscripts (n) and (n-1) refer to the current and previous time levels, and $\Delta\tau$ is the time step. The solution of the complete flow equations involves a global iteration process, in which the velocity-pressure coupling is effected by PISO-type predictor-corrector steps. The pressure equation is derived by introducing pseudo-velocities at staggered locations while maintaining the regular grid arrangement for all the transport equations. Fig.7(b) and (c) show the locations of nodes in the regular grid in the $\xi^2 \xi^3$ -plane. All transport quantities and pressure are evaluated at the regular nodes. In deriving the pressure equation, a control volume is employed as a continuity cell, to establish the coupling between the velocity and pressure fields. The pressure equation used in this study is written as

$$\begin{aligned}
& (E_d^{11} + E_u^{11} + E_n^{22} + E_s^{22} + E_e^{33} + E_w^{33}) p_P \\
& = E_d^{11} p_D + E_u^{11} p_U + E_n^{22} p_{NC} + E_s^{22} p_{SC} + E_e^{33} p_{EC} + E_w^{33} p_{WC} - \hat{D}
\end{aligned} \tag{26}$$

with

$$\hat{D} = \hat{U}_d^1 - \hat{U}_u^1 + \hat{U}_n^2 - \hat{U}_s^2 + \hat{U}_e^3 - \hat{U}_w^3$$

Here E^{ij} and a modified pseudovelocity \hat{U}^i at the regular node are

$$E^{ij} = \frac{R C_P}{J \left[1 + C_P \left(C_U + C_D + \frac{R}{\Delta\tau} \right) \right]} \sum_{m=1}^3 b_m^i b_m^j, \quad \hat{U}^i = \sum_{n=1}^3 b_n^i \hat{U}_n - E^{ij} \frac{\partial p}{\partial \xi^j} - E^{ik} \frac{\partial p}{\partial \xi^k} \tag{27}$$

where \hat{U}_i is a pseudovelocity given by the decomposition of Equation (25) for U_i into \hat{U}_i plus the pressure gradient terms, such that

$$U_i = \hat{U}_i - \frac{R C_P}{J \left[1 + C_P \left(C_U + C_D + \frac{R}{\Delta \tau} \right) \right]} \sum_{j=1}^3 b_i^j \frac{\partial p}{\partial \xi^j} \quad (28)$$

The coefficients and the modified pseudovelocities in the above equations are defined at the staggered node, and obtained from those at the regular node by the one-dimensional linear interpolation. The solution of the complete flow equations involves a global iteration process, in which the velocity-pressure coupling is effected by predictor-corrector steps. In the predictor step, the pressure field at the previous time step is used in the solution of the implicit equations (25) to obtain the corresponding velocity field. Since the velocity field generally does not satisfy mass conservation, a corrector step is needed. In the corrector step, the explicit momentum equations (28) and the implicit pressure equation (26) are solved iteratively to ensure the satisfaction of the continuity equation.

4.3 Multiblock (domain decomposition) capability, and overall numerical solution procedure

As mentioned earlier, the multiblock (domain decomposition) capability is facilitated in the present RANS method. This capability is essential for simulation of flow around complex geometry, e.g., multiple sail system for sailing yacht as focused in the present study. Fig.8 shows overview of the present multiblock computational grid, while the grid is generated by using an automatic gridding scheme developed by the present author (Masuyama et al., 2009) Note that the gridding engine together with the present RANS method was recently implemented into a comprehensive sail performance prediction software “*Advanced Aero Flow*” (Katori, 2009). See the reference for more details of the scheme. Total number of grids is around a half million, and the number of multiblock is 48. Free-stream, symmetry, and wall-surface (no slip) boundary conditions are imposed on outer and top boundaries, bottom boundary, and sail surface boundary, respectively. For the results shown in this paper, the mast and rigging are not considered in the series calculations, and the bottom boundary is located at the same height as that of deck plane of the boat (see Masuyama et al., 2009, for the results for which mast influences in computation are considered).

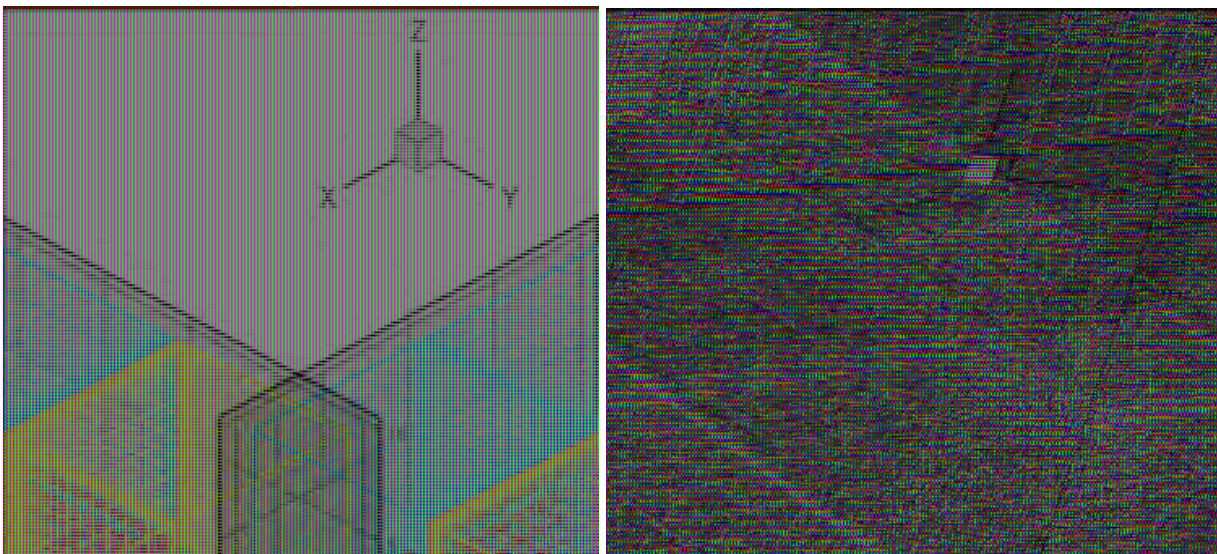


Fig. 8. Overview of the present multiblock computational grid.

The basic strategy to handle the multiblock follows domain decomposition technique to solve the elliptic PDE by using several subdomains. After adequate discretization is applied and a simple preconditioner is introduced, the discrete alternating Schwarz's method to solve the PDE is used for boundary matching. Finally, Fig.9 shows the code structure of the present RANS method, and the overall numerical solution procedure of the present RANS method is summarized as follows:

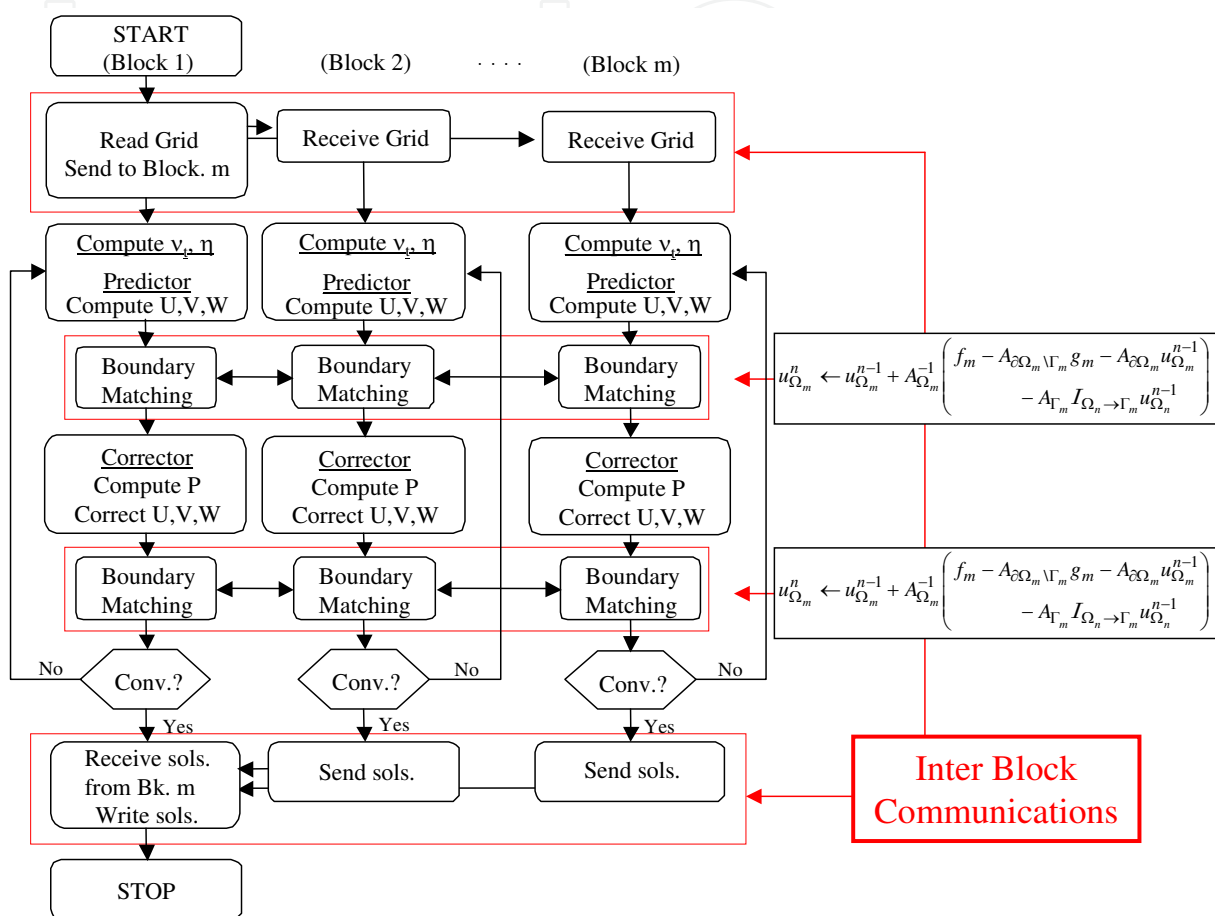


Fig. 9. PISO type solution algorithm for the present multiblock RANS method.

- Step 1. Input the computational grid, setup parameters, and boundary condition information.
- Step 2. Specify the initial conditions for the velocity, pressure and turbulence fields.
- Step 3. Compute the geometric coefficients.
- Step 4. Compute the finite-analytic coefficients for the transport equation.
- Step 5. Compute eddy viscosity distribution.
- Step 6. Solve transport equation for velocities (U, V, W) using the previous pressure field (predictor stage for the velocity field).
- Step 7. Compute the coefficients of pressure equation.
- Step 8. Solve pressure equation.
- Step 9. Using the newly obtained pressure, calculate the new velocity field explicitly (corrector stage for the velocity field).
- Step 10. Update the finite-analytic coefficients for the transport equation for velocities (U, V, W).

Step 11. Repeat Step 8 through Step 10 for the specified number of times.

Step 12. Return to Step 4 for the next time step, until the time step reaches the given maximum value.

More details of the present RANS method are described in Tahara et al. (2006a, 2006b) in addition to the above-cited references.

5. Measurements of upwind sail performance in full-scale condition using sail dynamometer boat *Fujin*

5.1 Full-scale measurements

Full-scale onboard measurements are free from scale-effect problem by wind tunnel tests and appear more promising, but the challenge becomes how to accurately measure forces acting on the sail. Such studies on sail force measurements were performed by Milgram et al. (1993), Masuyama et al. (1997a, 1997b), and Hochkirch et al. (1999), who built full-scale boats with onboard sail dynamometer systems.

Milgram (1993) showed in his pioneering work that the sail dynamometer boat, *Amphetrete*, is quite capable. This measurement system consists of a 35-foot boat with an internal frame connected to the hull by six load cells, which were configured to measure all forces and moments acting on the sails. In his work, the sail shapes were also measured and used for CFD analyses; however unfortunately, details of the sail shape and performance data were not presented. Hochkirch et al. (1999) also built a 33-foot dynamometer boat *DYNA*. The aerodynamic forces acting on the sail were measured and compared with the results from wind tunnel tests (Hansen et al. 2003). The measured data were also used as input to the CFD calculation and a parametric survey was carried out (Kreber et al. 2006). Masuyama and Fukasawa were encouraged by Milgram's work, and built a sail dynamometer boat, *Fujin*. The measurement system installed in the *Fujin* and the results of calibration test and sailing test were reported by Masuyama et al. (1997a and 1997b).

5.2 Measurements by sail dynamometer boat *Fujin*

The *Fujin* was originally built for conducting tests on sails for the Japanese America's Cup entry in 1994. *Fujin* is a 10.3m-long ocean cruiser with a sail dynamometer system in the hull which can directly measure sail forces and moments. Fig. 1 shows the general arrangement of the *Fujin*. The test sails were made to correspond to a typical sail plan for an International Measurement System (IMS) class boat. The rigging of the *Fujin* was originally designed for testing sails for the International America's Cup Class (IACC) boat. The jib of IACC boat is relatively small. Therefore, the longitudinal position of the jib rail track of the *Fujin* was located further forward than that of the typical IMS boat. For this reason, the tests were performed using a fully batten mainsail and a 130% jib instead of a 150% jib. The sails were made by North Sails Japan. The axes system is also shown in Fig. 1. The origin is located on the vessel's centerline at the aft face of the mast (x -direction), and the height of deck level at the base of the forestay (z -direction). Table 1 shows the principal dimensions of the boat and the detailed measurements of the sails, where "I", "J", "P" and "E" are the measurement lengths of sail dimensions for the IMS rule as defined in Fig.1.

The aerodynamic coefficients and the coordinates of the center of effort of the sails are defined as follows:

$$C_X = \frac{X_S}{\frac{1}{2}\rho_a U_A^2 S_A}, \quad C_Y = \frac{Y_S}{\frac{1}{2}\rho_a U_A^2 S_A}, \quad x_{CE} = \frac{N_S}{Y_S}, \quad z_{CE} = \frac{K_S}{Y_S} \quad (29)$$

where X_S and Y_S are the force components along the x and y axes of the boat respectively, and K_S and N_S are the moments around the x and z axes. x_{CE} and z_{CE} are the x and z coordinates of the center of effort of the sails (CE). The thrust force coefficient C_X is expressed as positive for the forward direction and the side force coefficient C_Y is positive for both port and starboard directions. It should be noted that the coordinates are given in the body axes system. Therefore, when the boat heels the Y_S force component is not in the horizontal plane but is normal to the mast. The aerodynamic forces acting on the mast and rigging are included in the measured sail forces.

5.2.1 Measurement system of aerodynamic performance and sail shape

The sail dynamometer system is composed of a rigid aluminum frame and four load cells. The frame is separated structurally from the hull and connected to it by the load cells. The general arrangement of the dynamometer frame is given in Fig.10. The load cells are numbered in the figure. Two of these are 1-component load cells and the others are 2-component ones. Hence, these load cells form a 6-component dynamometer system, and their outputs can be transformed to the forces and moments about the boat axes using a calibration matrix. All rig components such as mast, chain plates, winches, lead blocks, etc. are attached to the aluminum frame. The under deck portion of the mast is held by the frame, and the other rig components are attached to the frame through the deck holes. The data acquisition system and calibration method for the *Fujin* were described by Masuyama et al. (1997a and 1997b).

The sail shape was recorded using pairs of CCD cameras. The lower part of the mainsail was photographed using the CCD camera pair designated A in Fig.11. These were located at the mast top, 50 cm transversely from each side of the mast. The upper part of the mainsail was photographed using a portable video camera from below the boom. The lower part of the jib was photographed using the camera pair designated B in Fig.11, which were located at the intersection point of the forestay and the mast, 10 cm transversely from each side of the mast. The upper part of the jib was photographed using a portable video camera from inside the bow hatch. For measuring convenience, horizontal stripes were drawn on the mainsail and jib at heights of 10, 20, 40, 60 and 80% of each sail. The sail shape images were analyzed using the sail shape analyzing software, SSA-2D, developed by Armonicos Co. Fig.12(a) shows an example of processed image of the mainsail using the SSA-2D. This software calculates the curvature of the sail section by marking several points of the sail stripe and the reference line on the PC display, and indicates the parameters such as chord length, maximum draft, maximum draft position, entry angle at the luff, i.e., leading edge, and exit angle at the leech, i.e., trailing edge, as shown in Fig.12(b). The apparent wind speed (AWS) and apparent wind angle (AWA) are measured by an anemometer attached on the "Bow unit" as shown in Fig.11. This unit post can rotate freely to maintain its vertical attitude when the boat heels in order to measure the wind data in the horizontal plane. The height of the anemometer coincides with the geometric center of effort (GCE) of the sail plan. The wind speed and wind angle sensors were calibrated by wind tunnel tests in advance and the calibration equations were obtained. The *Fujin* also has motion measuring instruments such

as an Optical Fiber Gyroscope (roll and pitch angles), a Flux Gate Compass (heading angle), a Differential type GPS receiver, a speedometer (velocity in the x direction) and a potentiometer for rudder angle. These data are recorded by an onboard computer simultaneously with the data from the load cells.

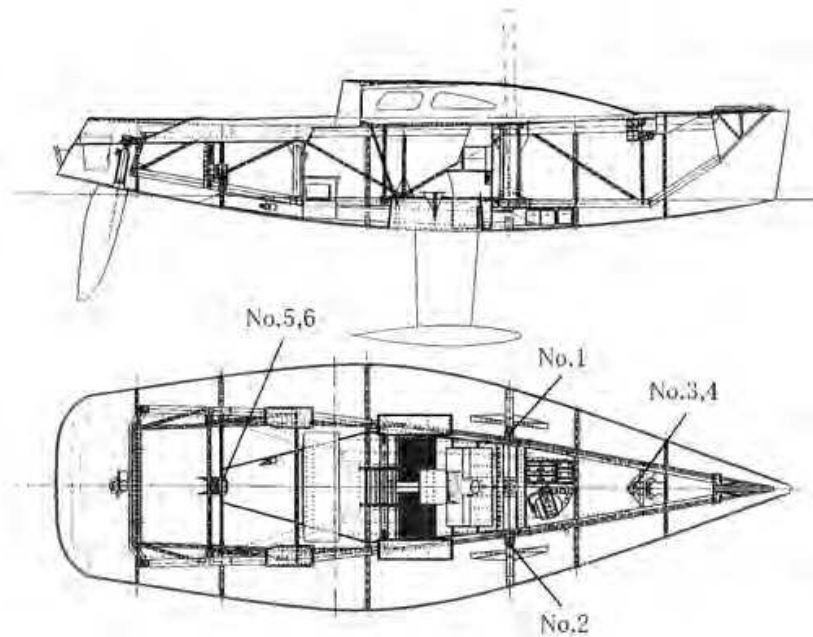


Fig. 10. General arrangement of dynamometer frame in *Fujin*.

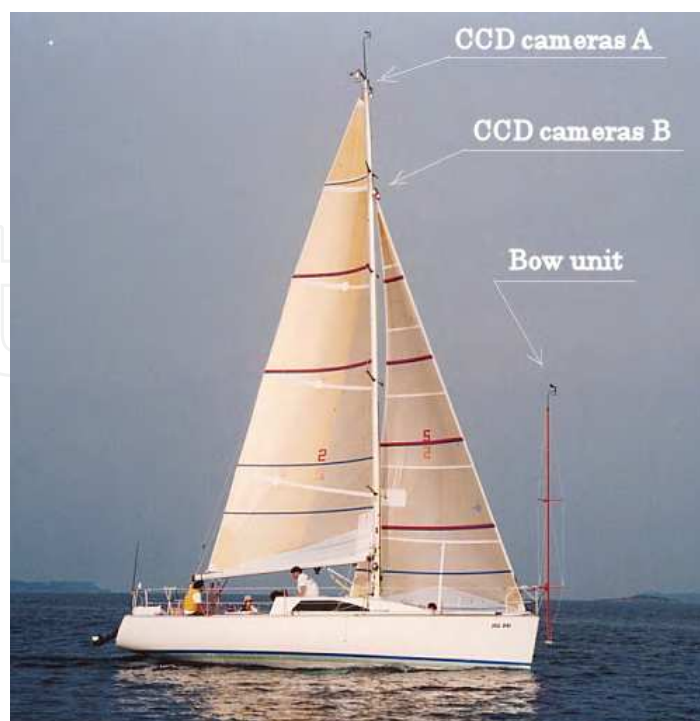


Fig. 11. Sea trial condition in light wind with 130% jib.

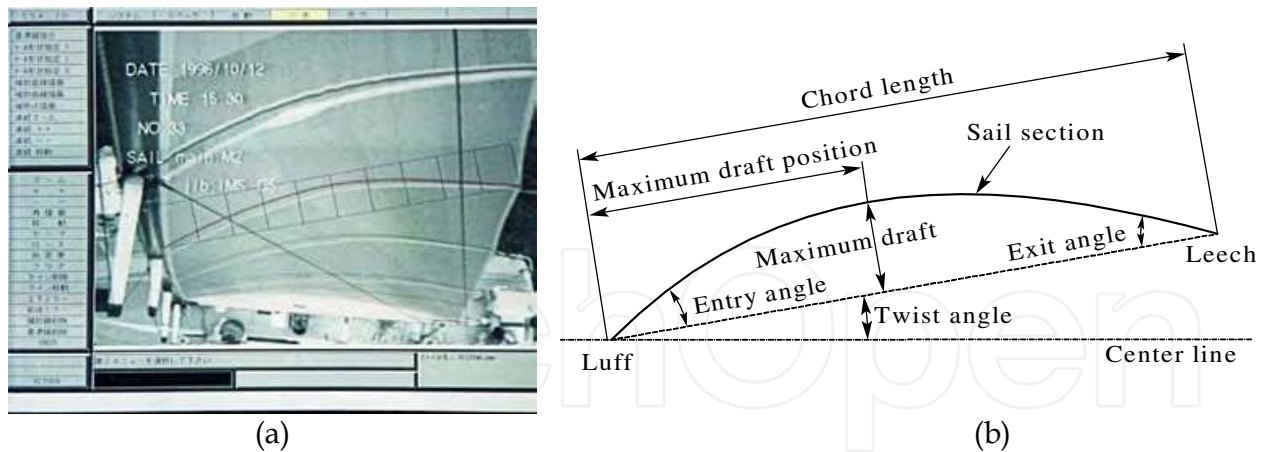


Fig. 12. (a) Example of a processed image of the mainsail using SSA-2D. (b) Measured sail shape parameters.

5.2.2 Test condition and error analysis

The sea tests were performed on Nanao Bay off the Noto Peninsula. The bay is approximately eight nautical miles from east to west and five from north to south. The bay is surrounded by low hills, and the mouth connecting it to the Japan Sea is narrow. Therefore, there is little tidal current in the bay, and the wave heights are relatively low even though the wind can be strong. The close-hauled tests were conducted over the apparent wind angle (AWA) range of 20 to 40 degrees, and the apparent wind speed (AWS) range of 5 to 11m/s. The effect of the AWA, and the draft and twist of the mainsail on the sail performance were measured. Data sampling was started when the sailing condition was considered to be in steady state. The sampling rate for the data acquisition system was set at 10Hz. Data sampling was continued for 90 seconds, and during this time the sail shapes were recorded using the CCD cameras. The boat was steered carefully during this time. However, the measured data contained some variation due to wind fluctuation and wave reflection on the hull. Therefore the steady state values for the aerodynamic coefficients were obtained by averaging the data over a 30 to 60 seconds period, in which the AWA was closer to the target value than during the whole 90 second period. For these tests if the range of deviation of AWA exceeded ± 5 degrees, the results were discarded. All of the measured coefficients are plotted with error bars indicating the range of deviation over the averaging period.

5.3 Comparison between experimental and calculated results

In this chapter, the experimental results and the calculated results for the following cases will be compared:

- a. Variation with apparent wind angle
- b. Variation with mainsail twist angle

For each series, first the sail coefficients: C_L , C_D , C_X and C_Y , and the coordinates of x_{CE} and z_{CE} are given. Then, the calculated the sail surface pressure and streamlines using the RANS-based CFD are presented for two typical cases in each series. Finally, the shapes and three-dimensional coordinates of the sails are tabulated for each case corresponding to those where the RANS-based CFD results are given.

5.3.1 Variation with apparent wind angle

Fig.13 shows the performance variation for the mainsail and 130% jib configuration as a function of AWA. In the figure the solid symbols indicate the experimental results and the open symbols indicate the calculated results using the VLM and the RANS-based CFD. For the experimental results, both data from the starboard (Stbd) and port tack (Port) are shown. All of the measured coefficients are plotted with error bars indicating the range of deviation over the averaging period. There are some discrepancies between the data from each tack. During the experiments, efforts were made to remove this asymmetrical performance. However, the boat speed actually differed on each tack. It can be concluded that there was a slight asymmetry in the combination of the hull, keel, rudder and dynamometer frame. The numerical calculations were performed using the measured shape data. In order to avoid confusion when interpreting the figure, the calculated results are indicated only for the port tack. Therefore, the calculated and experimental points for the port tack correspond to each other.

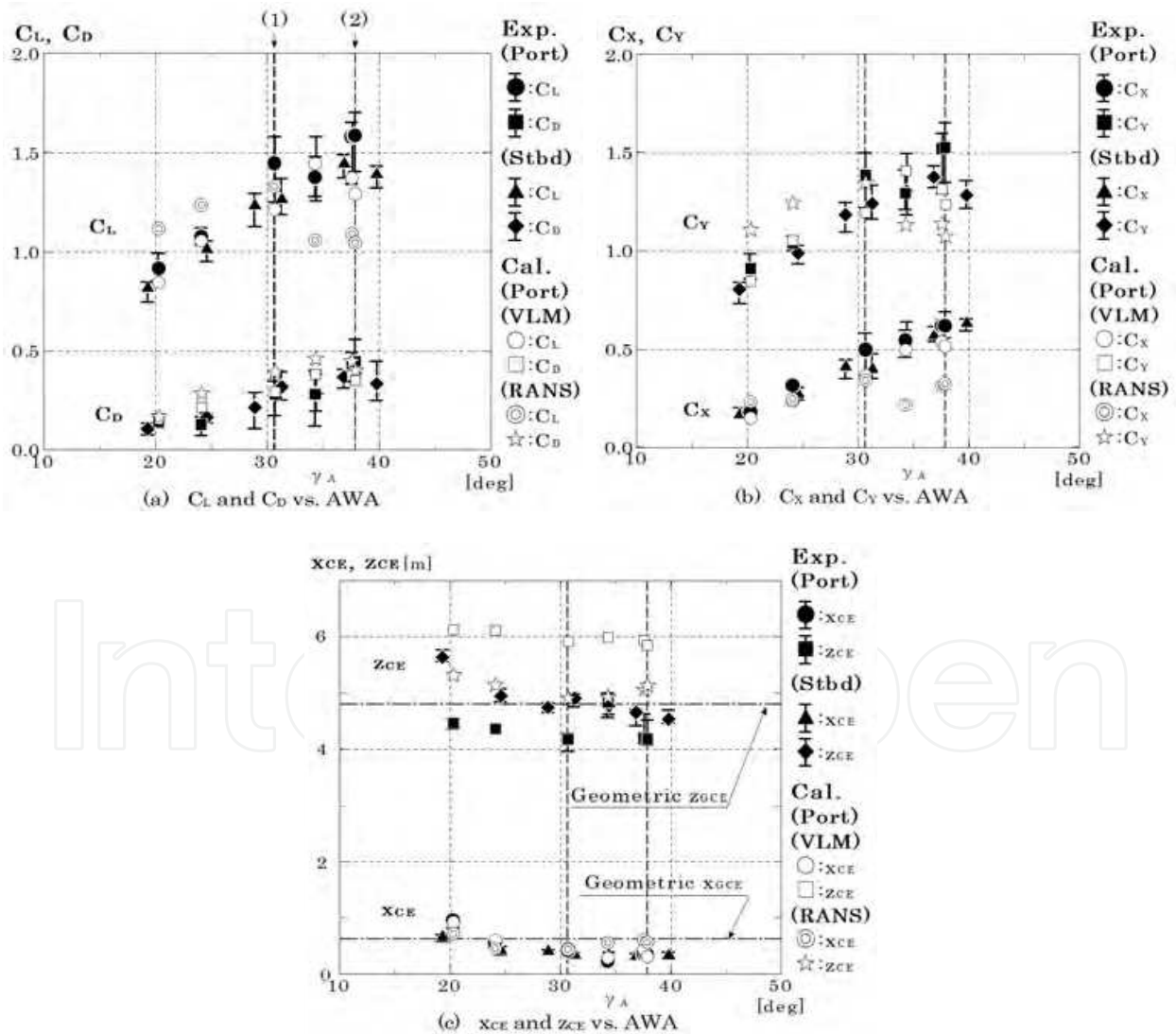


Fig. 13. Performance variation as a function of apparent wind angle (AWA) for mainsail and 130% jib.

In this figure, AWA ranges from 20.3 degrees to 37.9 degrees for the port tack. The former is the closest angle to the wind that was achieved, and the latter is typical of a close reaching condition, where the sail is eased for the power down mode. There is some scatter in the experimental data because this is made up from measurements taken with the sails trimmed in slightly different ways. The experimental kt value of C_L in Fig.13(a) varies with AWA from 0.91 to 1.58. For the close reaching condition, the sails were not well trimmed to satisfy the power down mode. A sample of measured sail sections at this condition is shown in a figure attached to Table 2(2). From the figure, it can be seen that both the mainsail and the jib are not eased sufficiently to correspond to the large AWA. This is the reason for the decrement in the measured lift curve slope of C_L at the range of AWA angles over about 35 degrees.

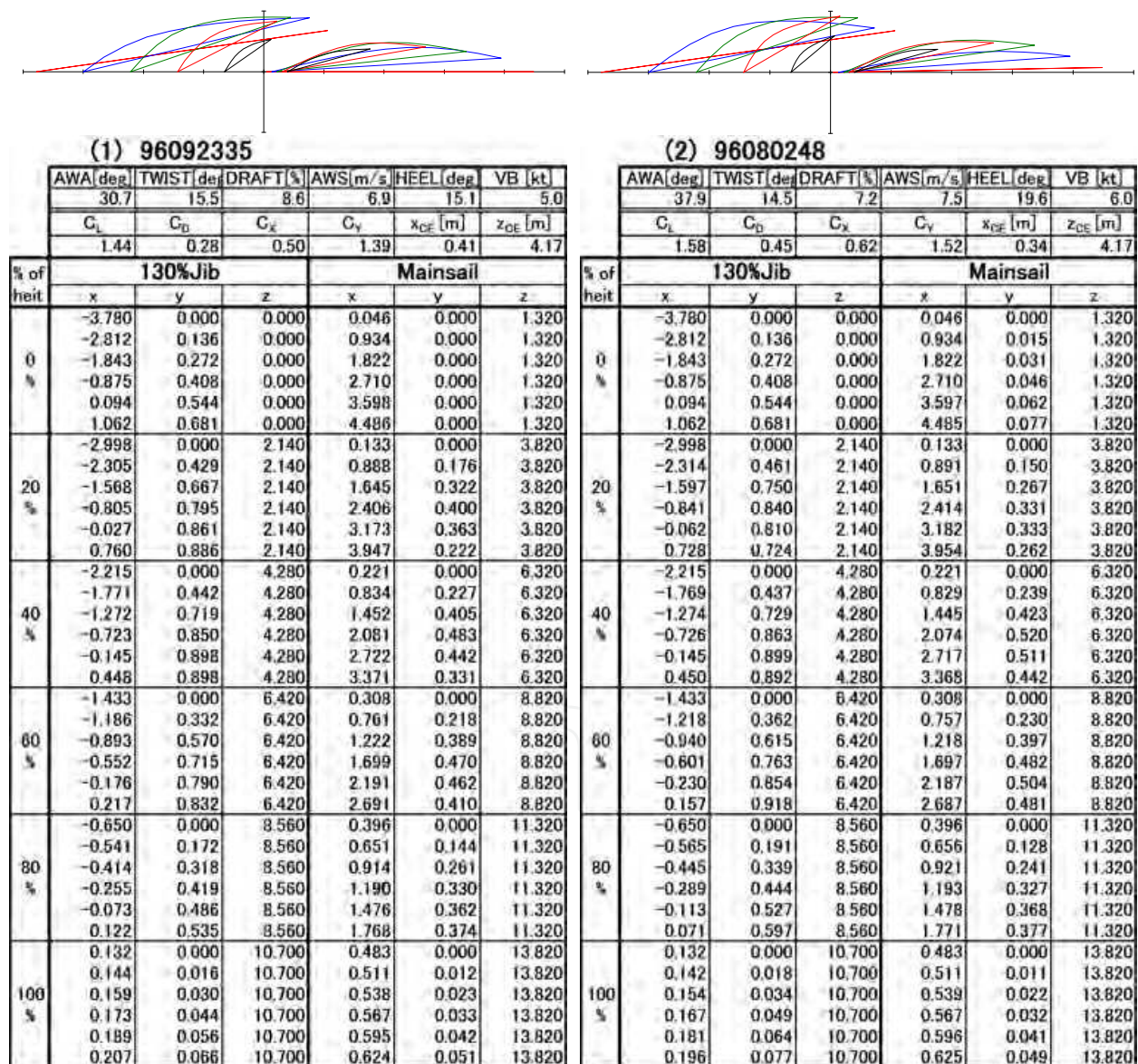
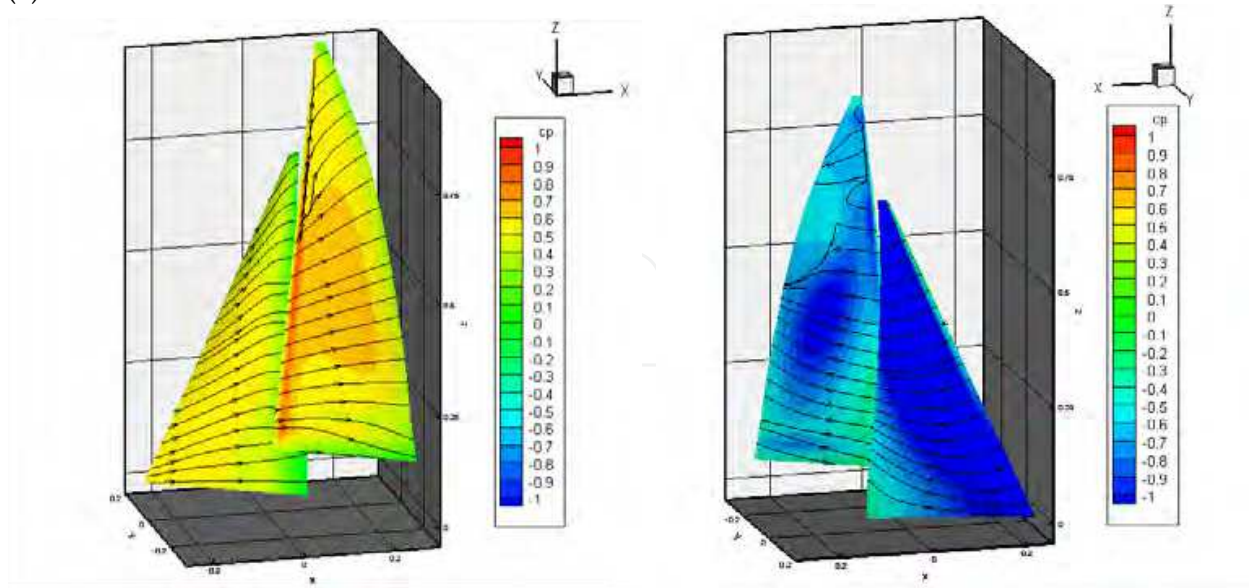


Table 2. Sail shapes, measured experimental data and three-dimensional coordinates of the sails for the cases of (1) 96092335 and (2) 96080248.

The calculated results for C_L using the VLM show good agreement with the experiments at AWA angles less than about 35 degrees. Over about 35 degrees, the calculated results are lower than the measured ones. This shows that the calculated results strongly indicate the effect of incorrect sail trimming. The results for C_L using the RANS-based CFD show the same trends with the experiments, but are slightly higher than those from the experiments for AWA between 20 degrees to 30 degrees and lower for AWA greater than 30 degrees. In particular, the decrease in C_L for AWA values greater than 30 degrees is considerably large. This will be discussed later with the calculated sail surface pressure and streamlines. The calculated results for C_D slightly over predict those from the experiments. Fig.13(c) shows the coordinates of the center of effort of the sails. The x and z coordinates of the geometric center of effort (x_{GCE} and z_{GCE}) are 0.63m aft and 4.80m above the origin, which are indicated by alternate long and short dashed lines in the figure. It is seen that both the experimental and the calculated coordinates of x_{CE} are near x_{GCE} and move slightly forward with increasing AWA. Unfortunately, there is a wide scatter in the experimental values of z_{CE} . This is thought to be because the measured K_s moment contains a large component from the mass of the dynamometer frame and rigging (659kg). This moment was subtracted from the measurement, taking into account the measured heel angle. If there is a slight error in the position of center of gravity of the dynamometer frame, or in the measured heel angle, the error in the calculated moment will be large. However, though there is a scatter in the measured data, it can be seen that z_{CE} is decreasing as AWA increases. The trends in the movement of both x_{CE} and z_{CE} as functions of AWA might be caused by the decrement of force acting on the aft and upper parts of the sails due to the loosening of main and jib sheets with increasing AWA. The calculated results for z_{CE} obtained using the RANS-based CFD show the same trend as the experiments. On the other hand, the calculated results using VLM are considerably higher than the experimental ones. This might be caused by over estimation of the force acting on the upper portion of the mainsail. In this area, since the jib is not overlapping, flow separation may occur easily. However, the VLM does not take flow separation into account.

Figures 14(1) and 14(2) show the calculated results of the sail surface pressure and streamlines using RANS-based CFD. Fig.14(1) indicates the case of experiment ID 96092335 (AWA= 30.7deg.), and 14(2) indicates ID 96080248 (AWA= 37.9deg.). These data correspond to the plotted points on the vertical dotted lines (1) and (2) in Fig.13. In Fig.14, the left and right diagrams correspond to the port and starboard sides, i.e., pressure and suction sides, respectively. In 14(1), although slight flow separation on the suction side of mainsail is seen, the streamlines of both sides run smoothly. On the other hand, in 14(2), considerable flow separation is occurring, in particular, on the suction side of jib. This is the main reason for the reduction of C_L value in the RANS-based CFD calculation at (2) in Fig.13(a). The shapes and three-dimensional coordinates of the sails are given in Table 2. The numbered (1) and (2) tables correspond to the cases of experiment ID 96092335 and ID 96080248, respectively. These also correspond to the calculated results shown in Fig.14. The figures described above the tables show the sail section profiles at 0, 20, 40, 60 and 80% of the sail height. The dimensions of these three-dimensional coordinates are given in the tables including 100% height section data. The positive direction of the x coordinate is aft. The four lines at the top of the tables are the measured values for the wind and sail trim conditions, the boat attitude and the sail performance coefficients.

(1)



(2)

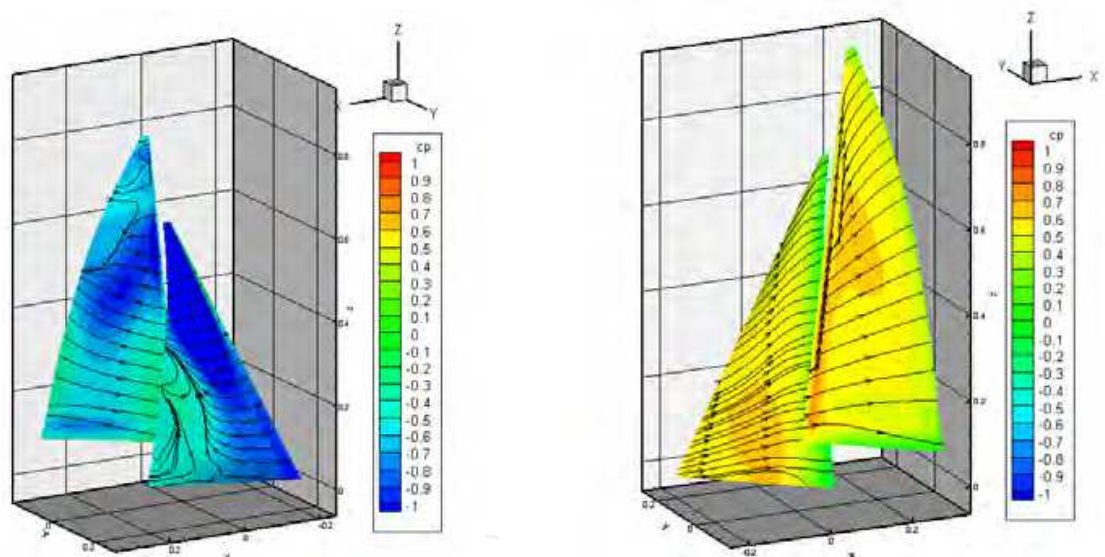


Fig. 14. (1) Surface pressure and streamlines obtained by RANS-based CFD at experimental ID 96092335 (AWA=30.7 deg.) and (2) ID 96080248 (AWA=37.9 deg.).

5.3.2 Variation with mainsail twist angle

Fig.15 shows the performance variation for the mainsail and 130% jib configuration as a function of mainsail twist angle. The mainsail twist was changed by varying the main sheet tension. The boom angle was kept parallel with the boat centerline by moving the main sheet traveler. The experiment was performed for an average value of AWA of 30 ± 2 degrees and mean draft at around 10%. The jib shape was fixed. The twist angle is defined as the angle between the boom line and section chord line at 80% height. In the figure, the

twist angle ranges from 4.5 degrees to 24.9 degrees for the port tack. Varying the twist angle by 20.4 degrees, results in the value of C_X in Fig. 15(b) changing from 0.33 to 0.39 (18%), and the value of C_Y changing from 1.16 to 1.39 (20%). It can be seen that the maximum C_X occurs at a twist angle of around 15 degrees. The considerable decrease in C_Y with increasing twist angle is also worth noticing. In this case, the calculated results of both VLM and RANS-based CFD for C_X and C_Y , and C_L and C_D correspond to the measured values very well.

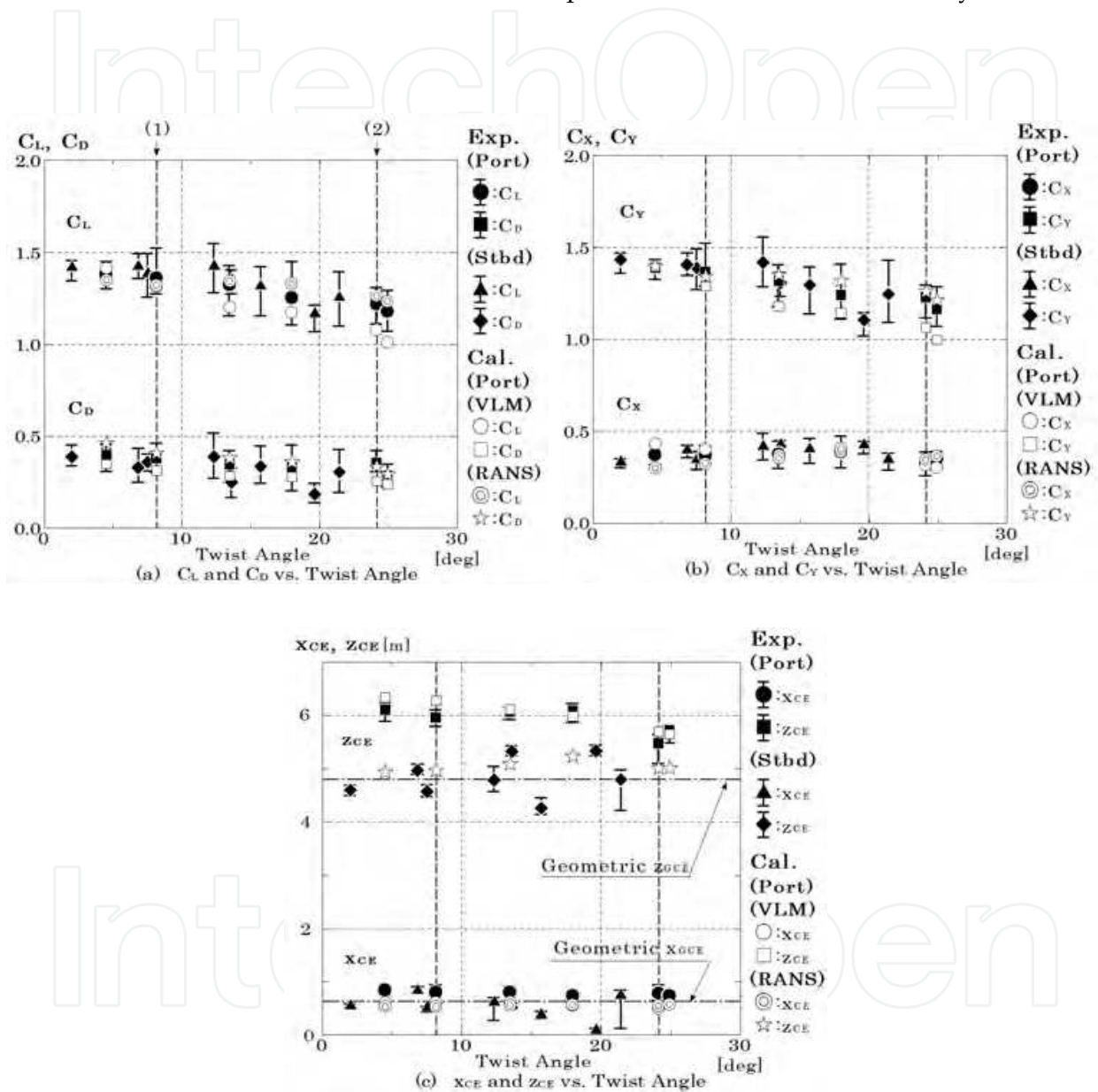


Fig. 15. Performance variation as a function of mainsail twist angle for mainsail and 130% jib.

Figures 16(1) and 16(2) show the calculated results using RANS-based CFD. Fig.16(1) corresponds to ID 97072213 (twist angle = 8.2 deg.), and 16(2) corresponds to ID 97072218 (twist angle = 24.1 deg.). It can be seen in Fig.16(1) that the streamlines on the upper part of the suction side of the mainsail for the smaller twist angle, show considerable flow

separation. This is caused by the large angle of attack at the upper part of the sail due to the small twist angle. On the other hand, for the higher twist angle shown in Fig.16(2), there is a low negative pressure area at the luff on the suction side of mainsail due to the small angle of attack. This is what causes the considerable reduction in the calculated value for C_x at (2) in Fig.15(b). Table 3 shows the shapes and three-dimensional coordinates of the sails for cases (1) and (2), which correspond to the calculated results shown in Fig.16. Further measured data and comparison with the numerical calculations are described by Masuyama et al. (2007 and 2009)

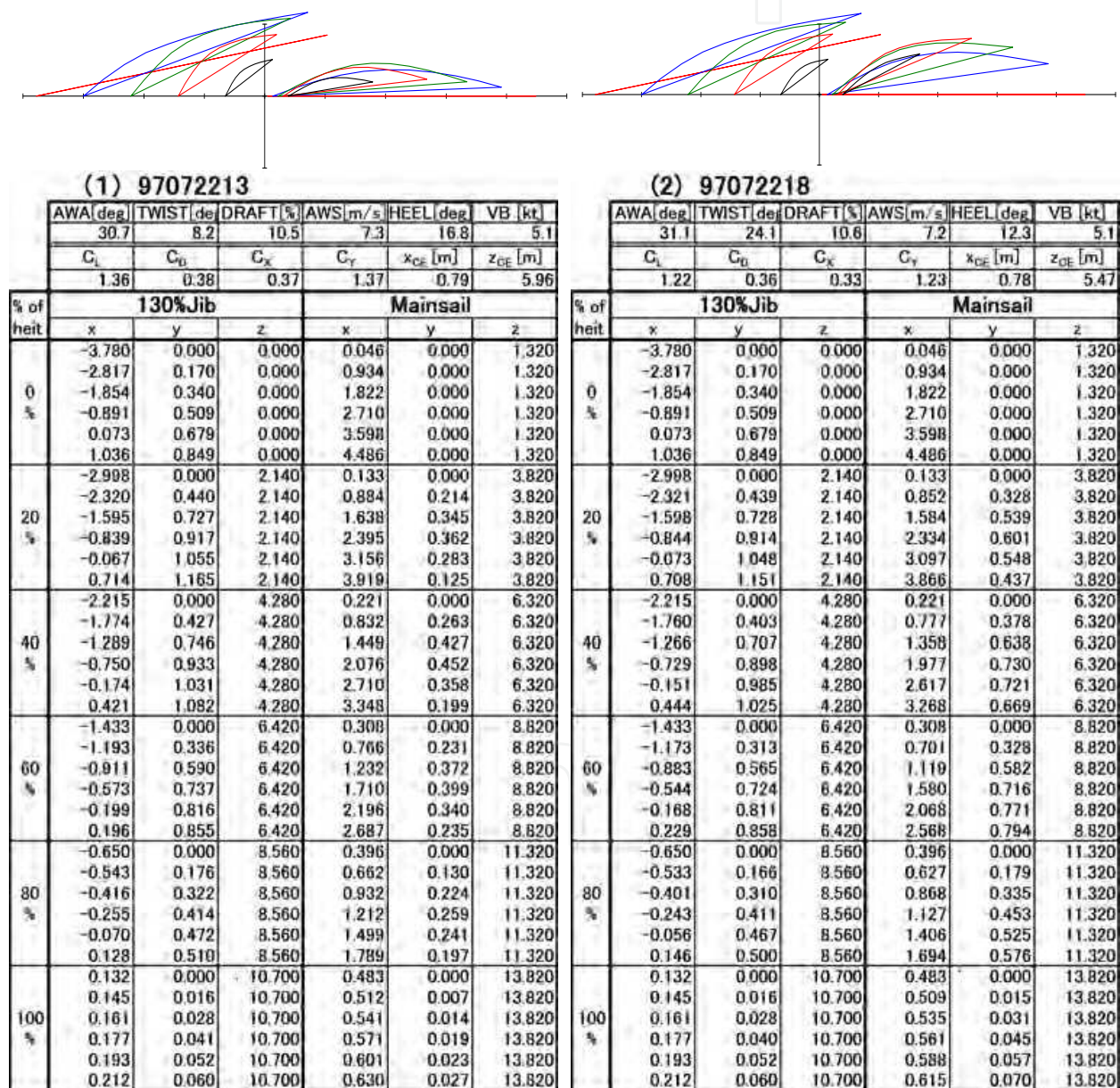
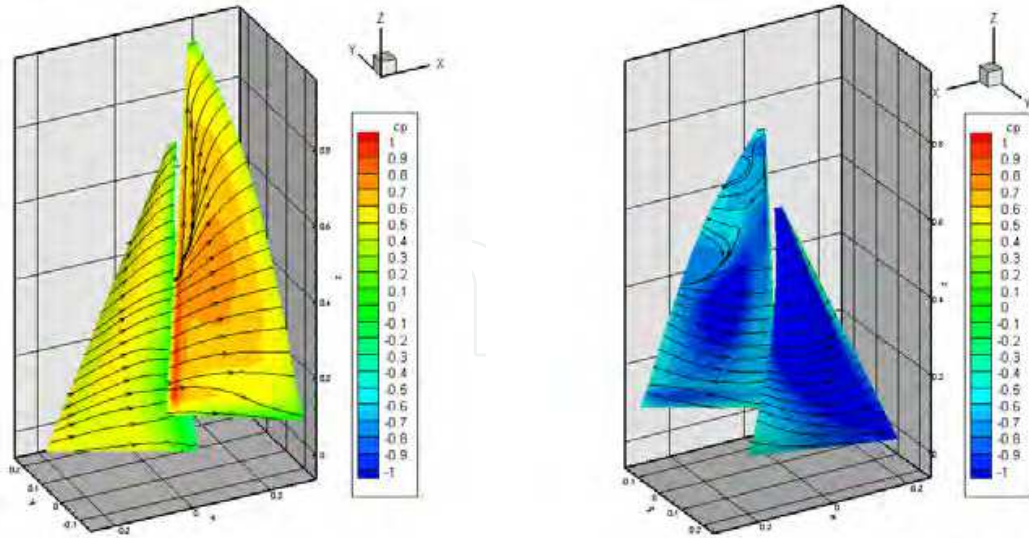


Table 3. Sail shapes, measured experimental data and three-dimensional coordinates of the sails for the cases of (1) 97072213 and (2) 97072218.

(1)



(2)

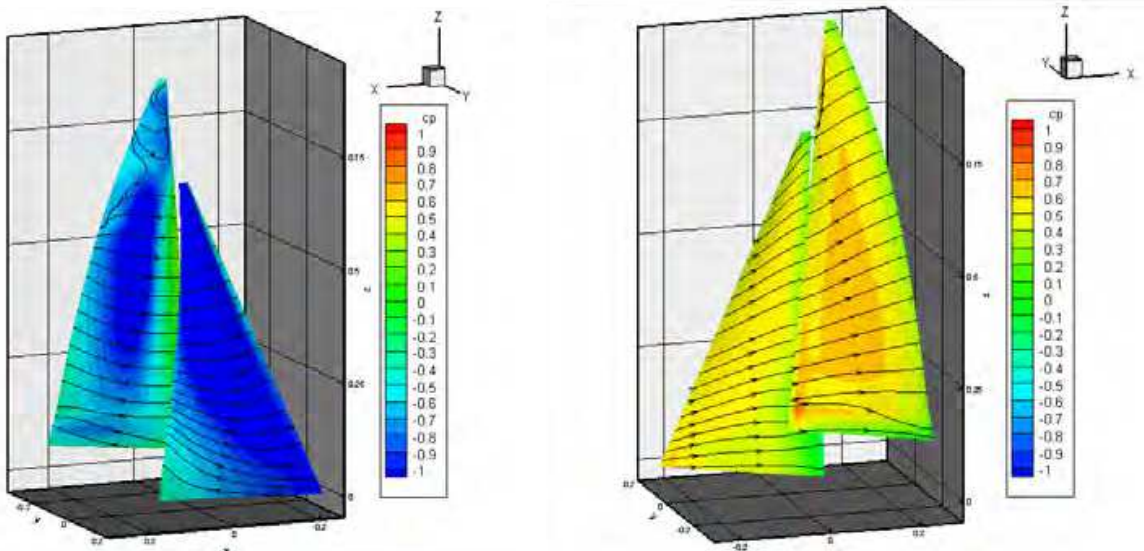


Fig. 16. (1) Surface pressure and streamlines obtained by RANS-based CFD at experimental ID 97072213 (twist angle= 8.2 deg.) and (2) ID 97072218 (twist angle= 24.1 deg.).

6. Discussion of numerical calculation methods

The flow is dominated by multiple-lifting-surface aerodynamic interactions. For larger AWA values, in particular, a large-scale flow separation exists on the leeward side of the sails. In general, there is complex vortex generation in the wake, especially near the top and bottom of the sails, i.e., tip vortices are generated and are influenced by the boundary layer flows on the sails. The resultant aerodynamic forces are mostly dominated by the pressure component, whereas the contribution of the frictional component is generally small. The accurate prediction of the boundary layer flows on the sails and the three-dimensional flow separation, associated with the abovementioned vortex generation, are big challenges for

RANS-based CFD. The geometrical complexity is also another significant challenge to RANS-based CFD. The accuracy in the prediction of the CE is of great interest, in association with the correct prediction of the above-mentioned three-dimensional flow separation.

Through the analyses of the multiblock RANS-based CFD, it appears that the overall trends of the flow and the aerodynamic forces measured in the experiments are fairly well predicted by the present computations. It is also seen that the multiblock domain decomposition considered here is very effective for the present mainsail and jib configurations. The automatic gridding scheme used successfully generates high-quality structured grids for the various sail geometries, AWA, and heel angles considered in the present study. Although there are advantages to a structured grid system for high-resolution in the boundary layer flow, building a grid in this fashion is difficult to apply to complex geometries. This problem appears to be resolved by the present scheme.

The Vortex Lattice method is, on the other hand, a convenient tool to predict the lift and induced drag acting on the sail accurately at the apparent wind angles less than about 35 degrees. The computational time of the method is about a few minutes for one calculation condition. The longitudinal coordinates, or x coordinate, of the center of effort of the sail can also be calculated with accuracy by the Vortex Lattice method, however, the estimated vertical coordinate, or z coordinate, of the center of effort by the Vortex Lattice method is considerably higher than the experimental ones. This may be caused by the fact that the flow at the upper portion of mainsail is easily separated because of the absence of jib overlapping, while the flow separation cannot be taken into account in the Vortex Lattice method

7. Conclusion

The sail performance analysis of sailing yacht was carried out by using numerical calculations and experiments. Focus in the present manuscript is especially on the upwind sailing condition. The sails considered here are IMS type, and the shapes and performance were measured by using the sail dynamometer boat *Fujin*. The measured sail flying shapes were used by numerical analysis, where two CFD methods developed by the authors were used, i.e. a multiblock RANS-based CFD method by Tahara and a VLM-based CFD method by Fukasawa. It appears that the overall trends of the flow and the aerodynamic forces measured in the experiments are fairly well predicted by the present computations; and at the same time, the present sail performance database based on the full scale onboard measurements are very useful for validation study of numerical methods. As compared to maturity of VLM, that for RANS-based CFD is still in underway but the future prospect is shown promising, especially for capability in predicting separation flow field where viscous effects of fluid are significant. The authors believe that our sail performance database associated with accurate sail flying shape measurements will be able to contribute to the further development of more advanced CFD methods.

Although details are not described in the present manuscript, our current effort is directed toward the more challenging problem, i.e., extension of the present work for the downwind sailing condition. Since the onboard sail shape measurement system of *Fujin* is incompetent for the spinnaker measurement due to its balloon shape, the sail shapes and performance are measured using wind tunnel equipment, and such activities are already in progress. The sail shapes are recorded using digital cameras and processed to obtain 3D coordinates using solid shape analyzer software, which provides 3D coordinates from digital photographs taken from several different directions. Importantly, the sail shapes and the sail forces and

moment acting on the model are measured simultaneously. The numerical simulation by using a RANS-based CFD method is also in progress. Along with integration with the aforementioned sail design and performance prediction software *AAF*, more details of our work will be reported in our future publications.

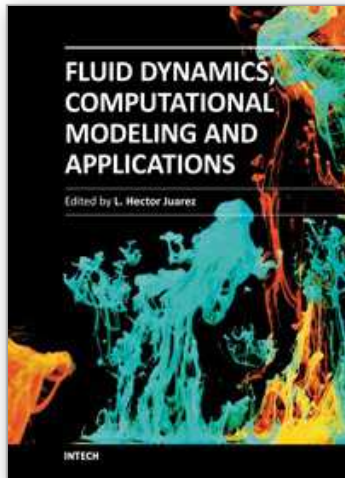
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