The Space Congress® Proceedings 2018 (45th) The Next Great Steps

Feb 28th, 7:30 AM - 8:30 AM

Spacecraft Radiation Shielding by a Dispersed Magnetic Field Array

Dr. David Chesny
Postdoctoral First Award Fellow, National Space Biomedical Research Institute

S. T. Durrance
Florida Institute of Technology

G. A. Levin
Florida Institute of Technology

Scholarly Commons Citation

Chesny, Dr. David; Durrance, S. T.; and Levin, G. A., "Spacecraft Radiation Shielding by a Dispersed Magnetic Field Array" (2018). *The Space Congress® Proceedings*. 7.
https://commons.erau.edu/space-congress-proceedings/proceedings-2018-45th/feb-28-2018/7

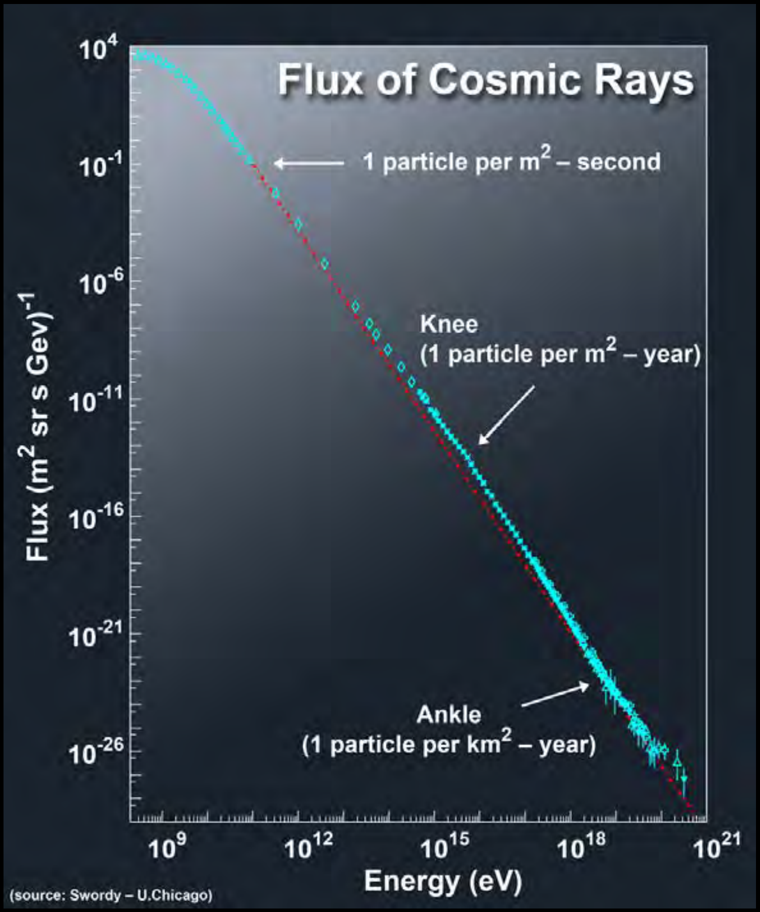
Spacecraft Radiation Shielding Using Dispersed Superconducting Loops

D. L. Chesny[1], S. T. Durrance[2], G. A. Levin[2]

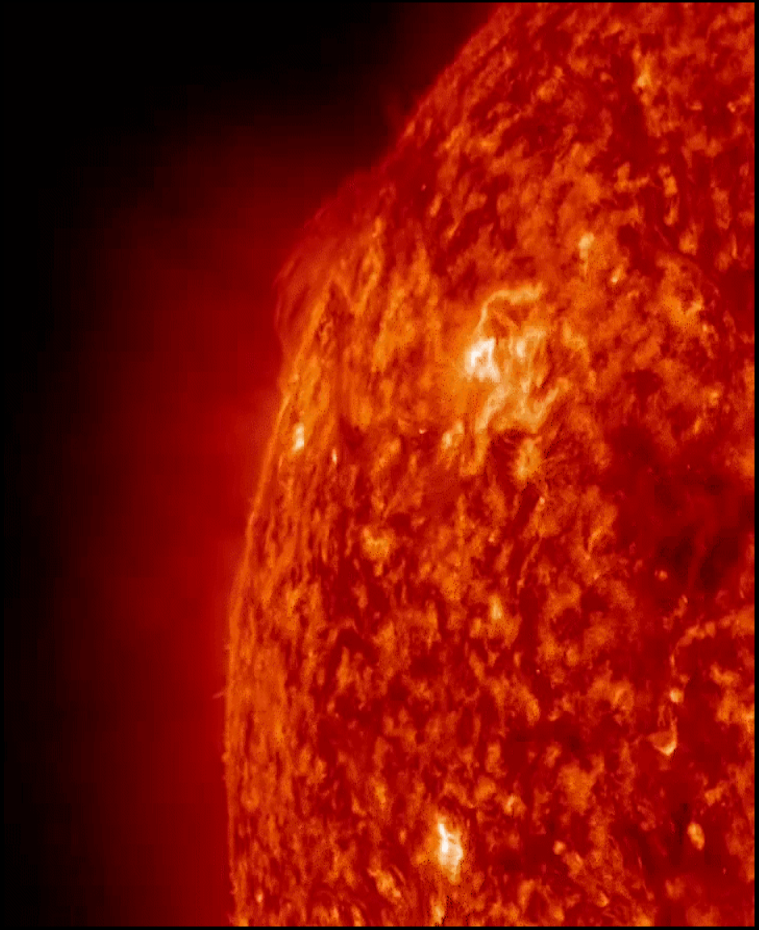
[1]OrangeWave Innovative Science, LLC

[2]Florida Institute of Technology

Interplanetary Radiation Environment



(source: Swordy – U.Chicago)



Galactic Cosmic Rays (GCRs)

Isotropic and *constant*
1—1000 GeV
Protons ←→ Fe

Solar Particle Events (SPEs)

Isotropic and *intermittent*
1—1000 MeV
protons, H, He, C, Si, Fe

Radiation Threat

Radiation Exposure Induced Death (REID)

NASA Standard of **<3%** increase (95% confidence)

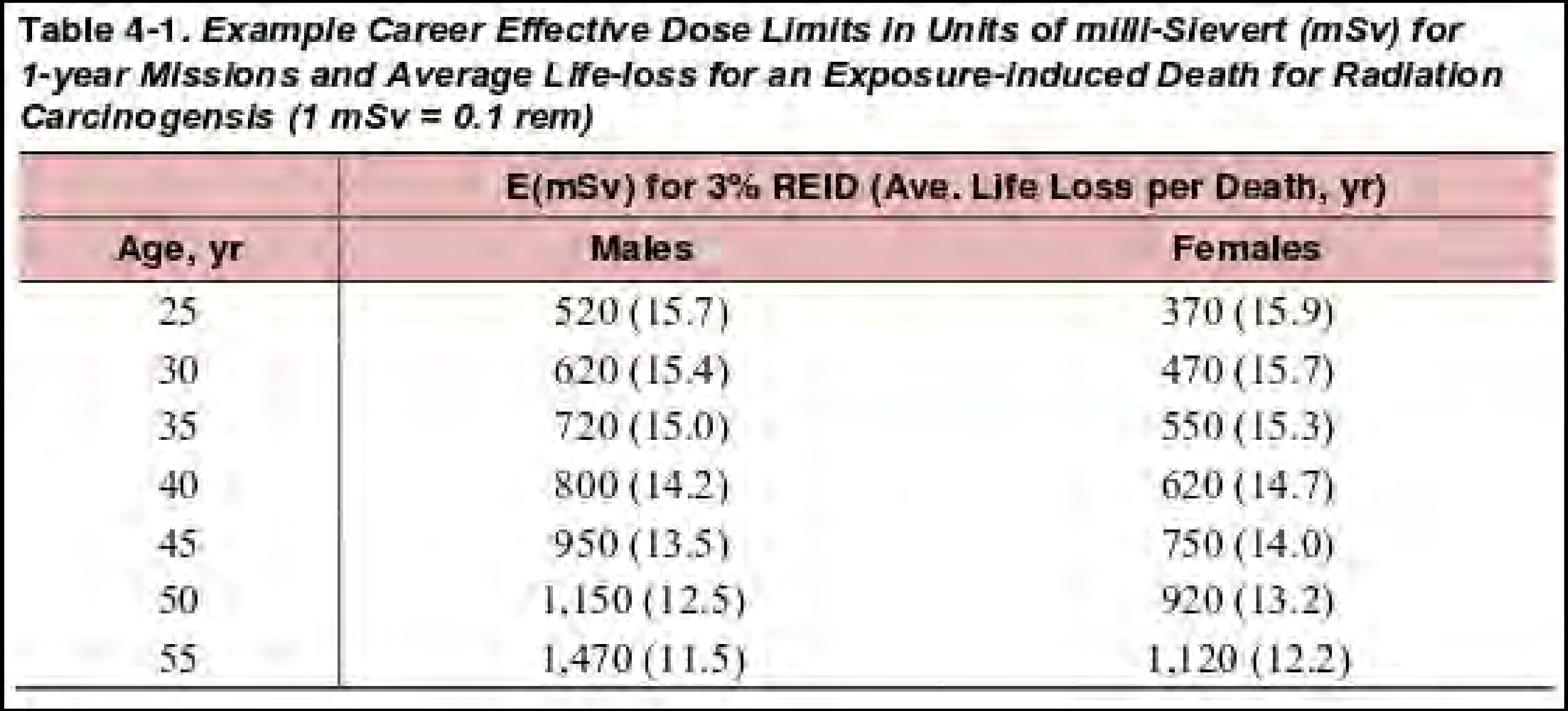
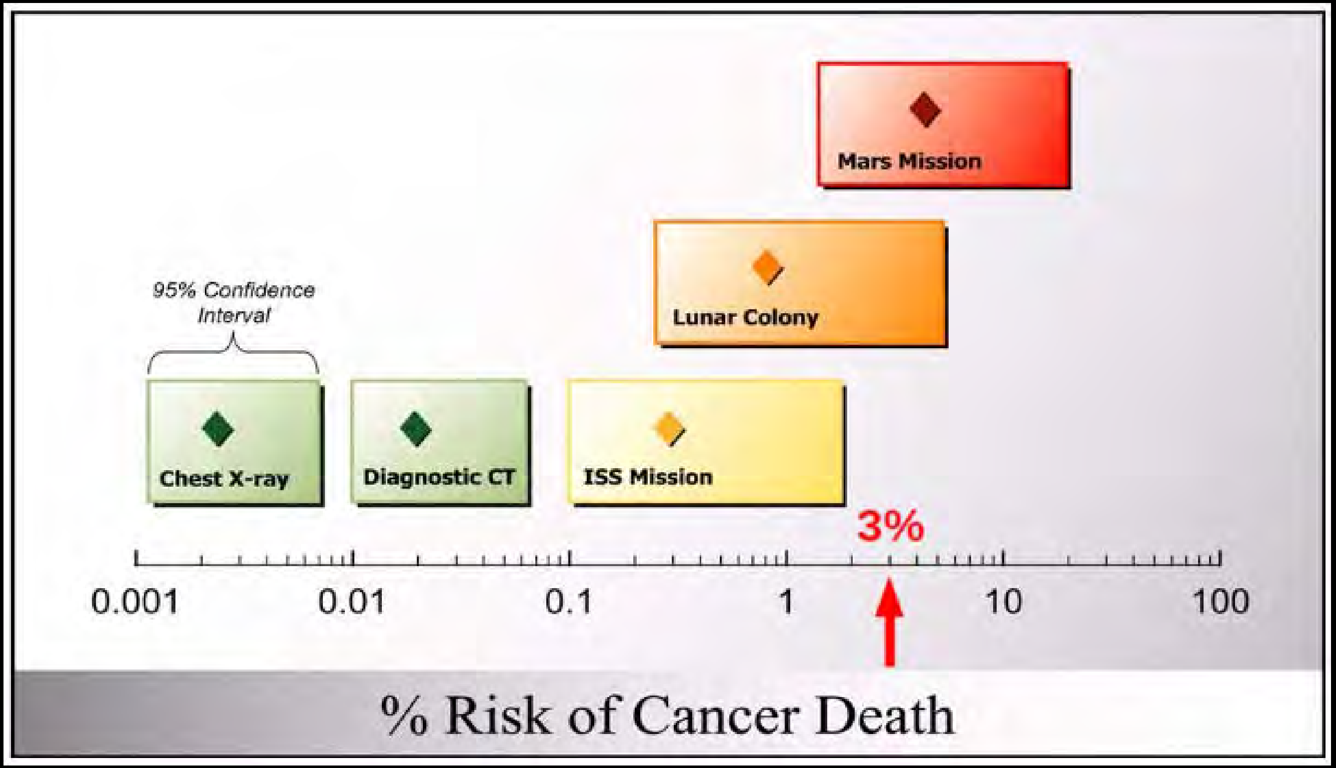


Table 4-1. *Example Career Effective Dose Limits in Units of milli-Sievert (mSv) for 1-year Missions and Average Life-loss for an Exposure-Induced Death for Radiation Carcinogensis (1 mSv = 0.1 rem)*

	E(mSv) for 3% REID (Ave. Life Loss per Death, yr)	
Age, yr	**Males**	**Females**
25	520 (15.7)	370 (15.9)
30	620 (15.4)	470 (15.7)
35	720 (15.0)	550 (15.3)
40	800 (14.2)	620 (14.7)
45	950 (13.5)	750 (14.0)
50	1,150 (12.5)	920 (13.2)
55	1,470 (11.5)	1,120 (12.2)

Radiation Threat



“The NASA **PELs** for fatal cancer risk may be exceeded for several lunar scenarios including a large SPE, cumulative career exposure, and mission length dependent on crew age and gender. In addition, the NASA PELs for fatal cancer risk are projected to be <u>violated under all possible Mars scenarios</u> at this time.”

-NASA Human Research Roadmap

Curiosity Rover

253-day cruise to Mars

SPEs

GCRs

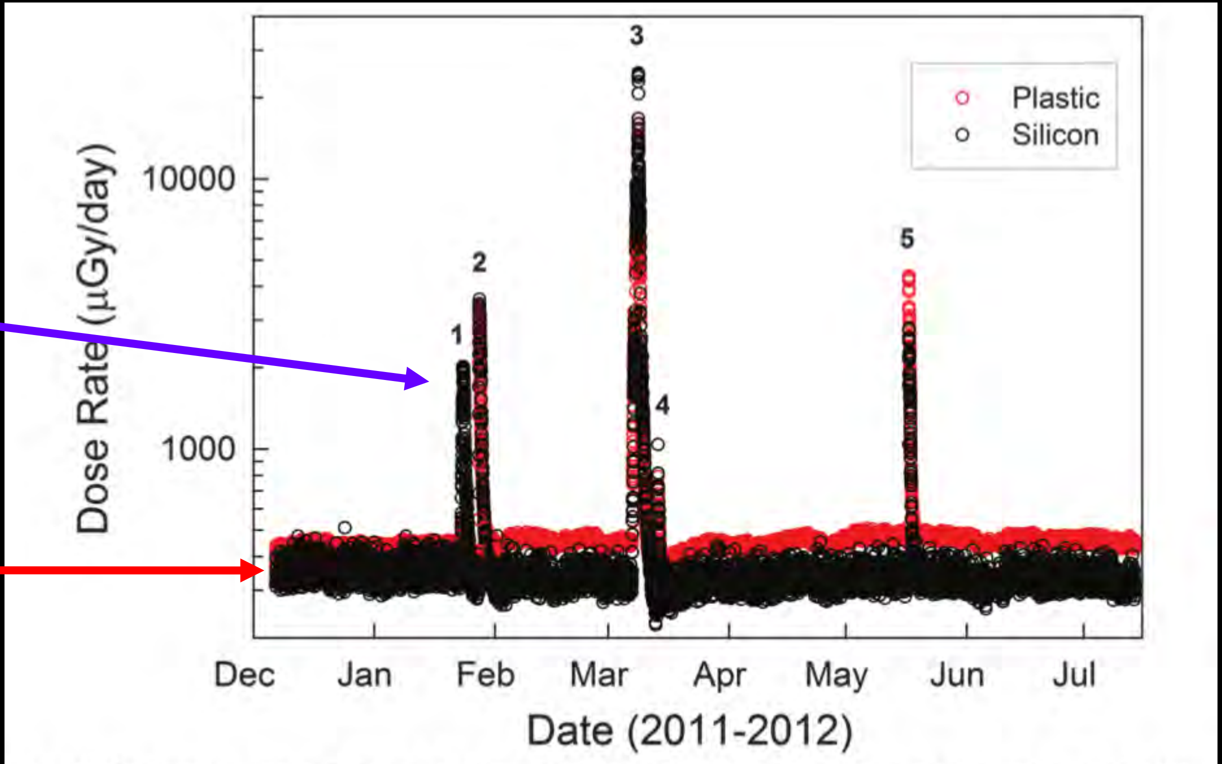


Fig. 1. Dose rates recorded in a silicon detector (black circles) and in a plastic scintillator (red circles) during the MSL's cruise to Mars.

Shortest round-trip: 660±120 mSv

Why magnetic shielding?

Threat Mitigation

Deal with consequences

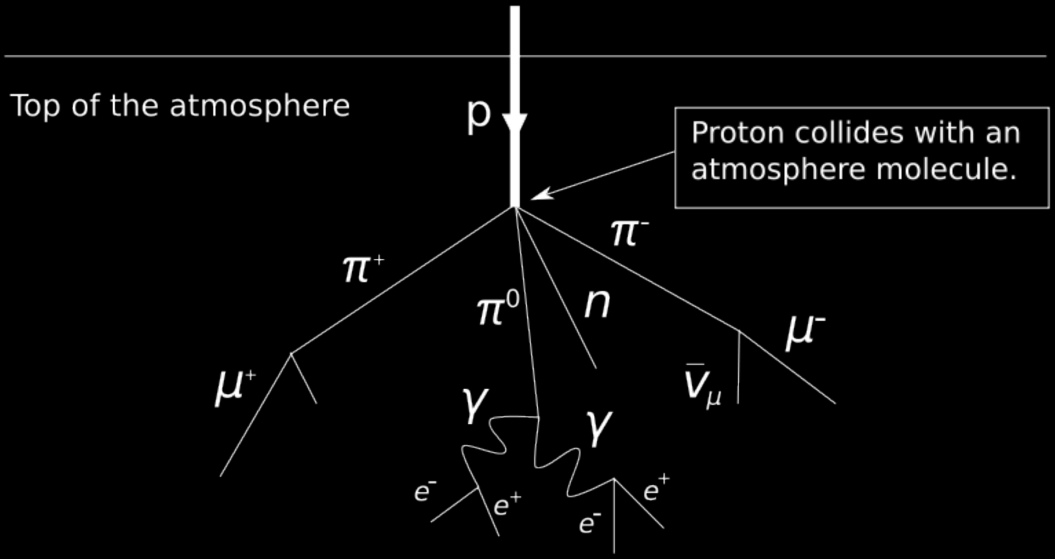
- Molecular/DNA level
- Enterade
- Not a “showstopper”

Prevention

- Absorption
- Deflect

Absorption - Material Shielding

Secondary particles *increase* exposure



Proton collides with an atmosphere molecule.

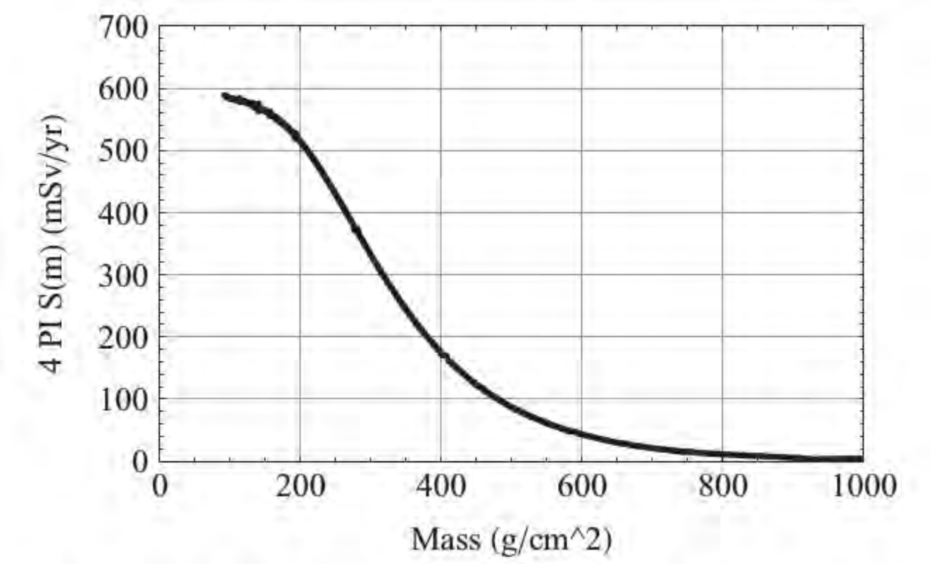
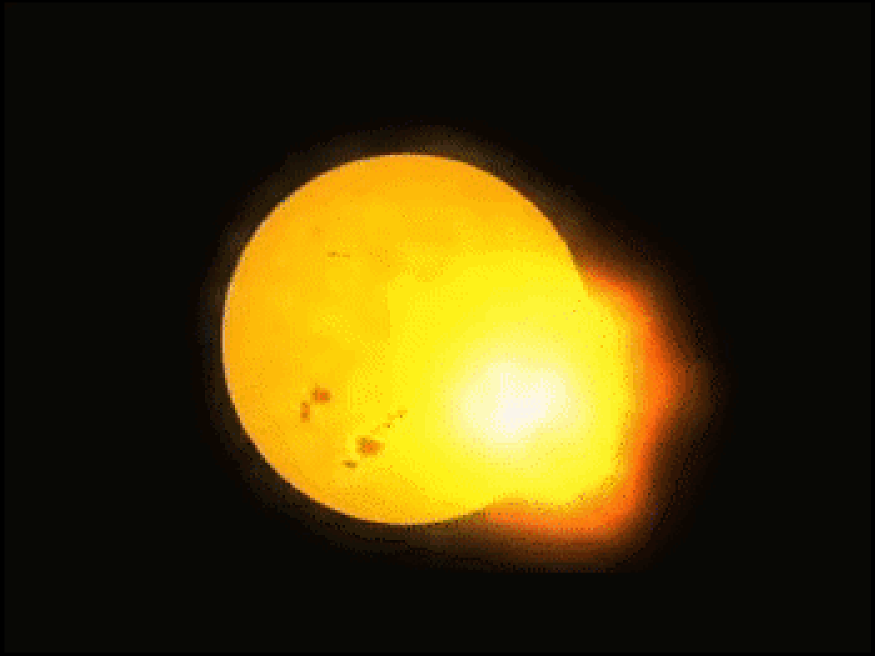
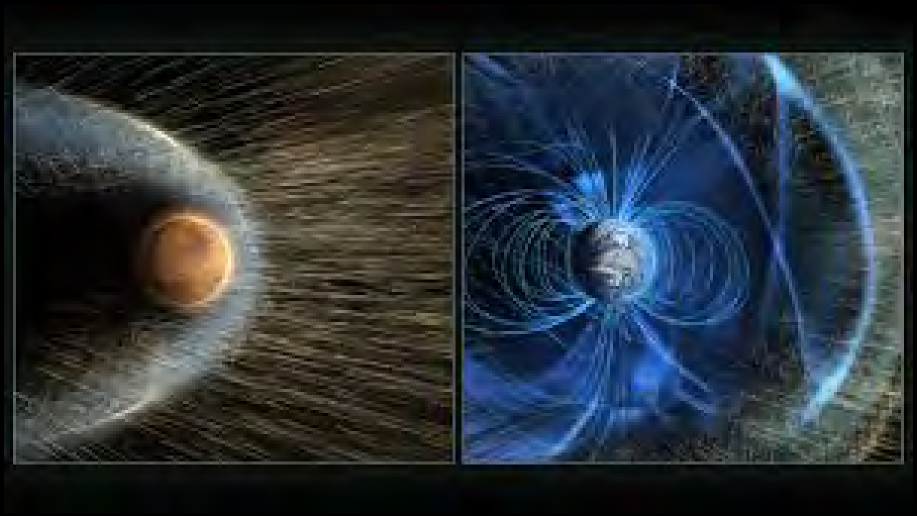


Fig. 5. This plot shows the atmospheric radiation shielding function at the 06/09 solar minimum,

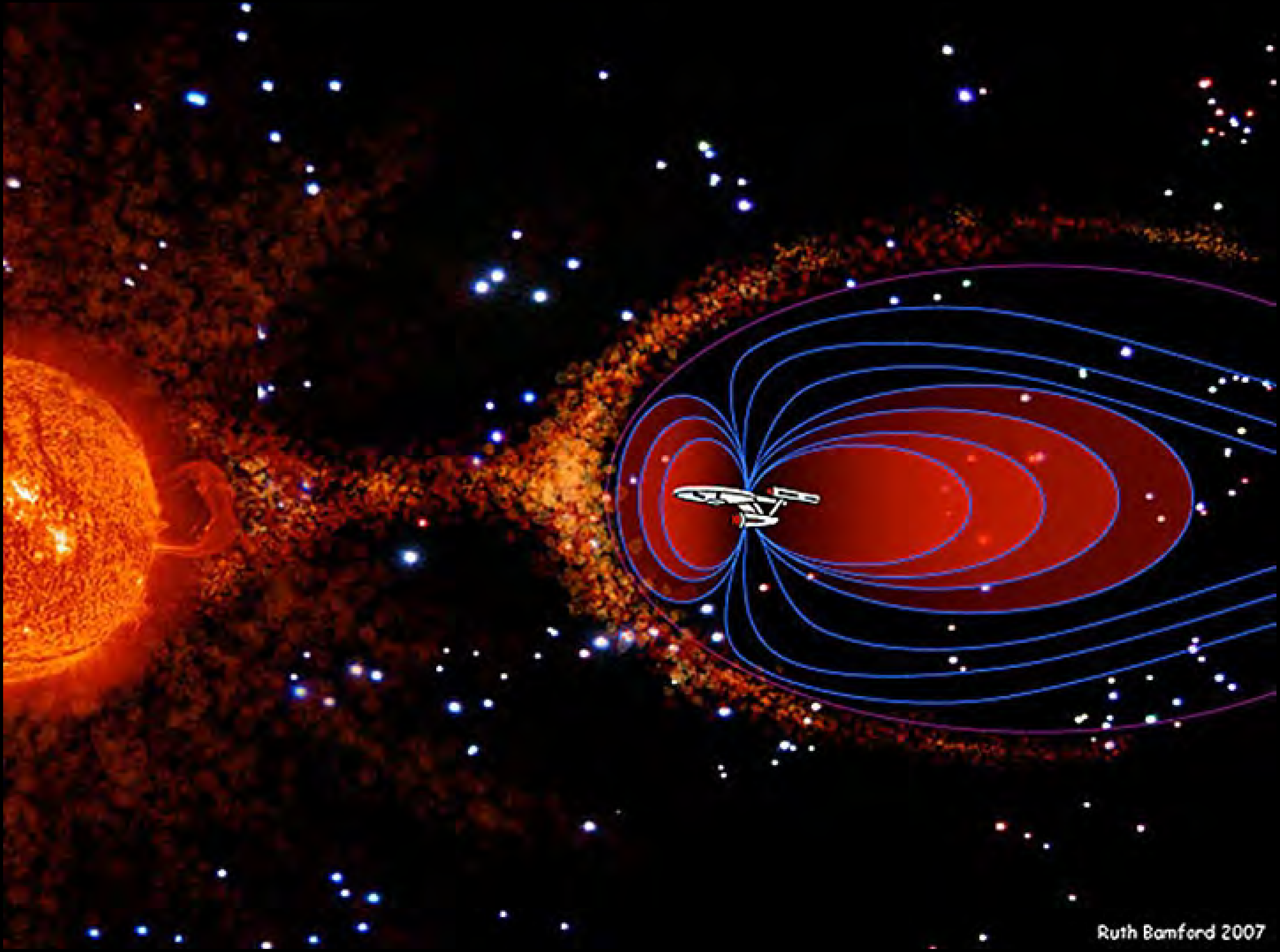
Youngquist et al. (2014)

Deflection - Mimic Earth's Shield

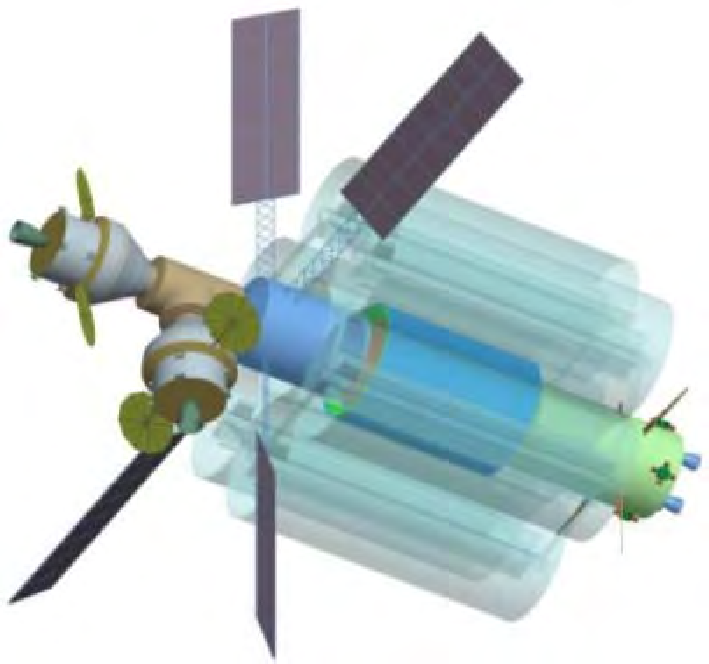




Magnetic Shielding

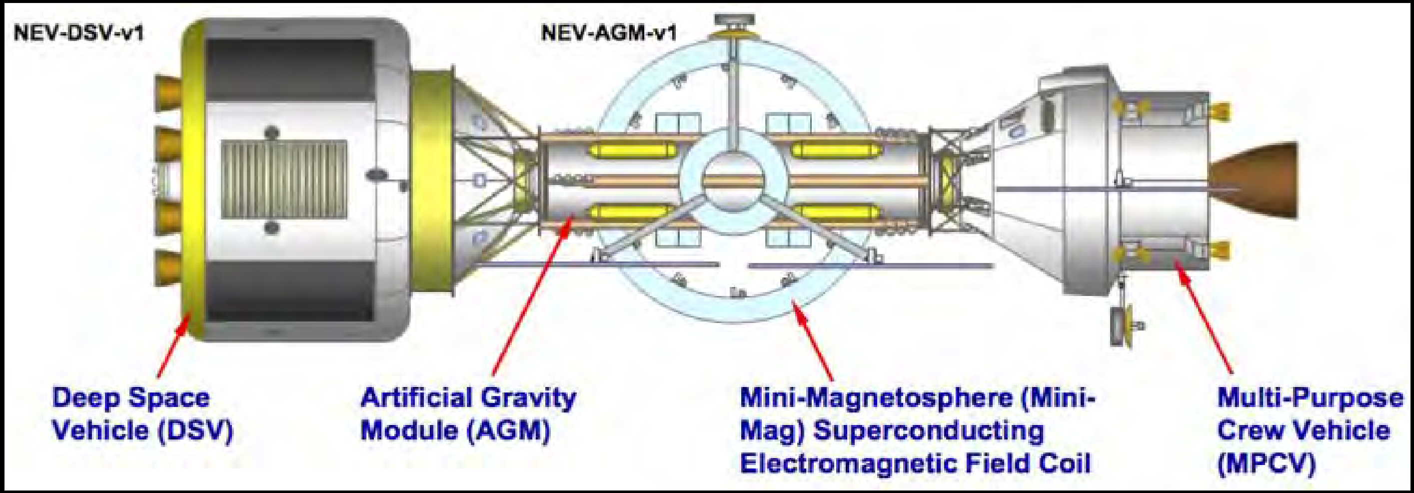


Previous Designs



Kervendal, E., Kirk, D., Meinke, R. (2006)

Superconducting magnets attached directly to spacecraft

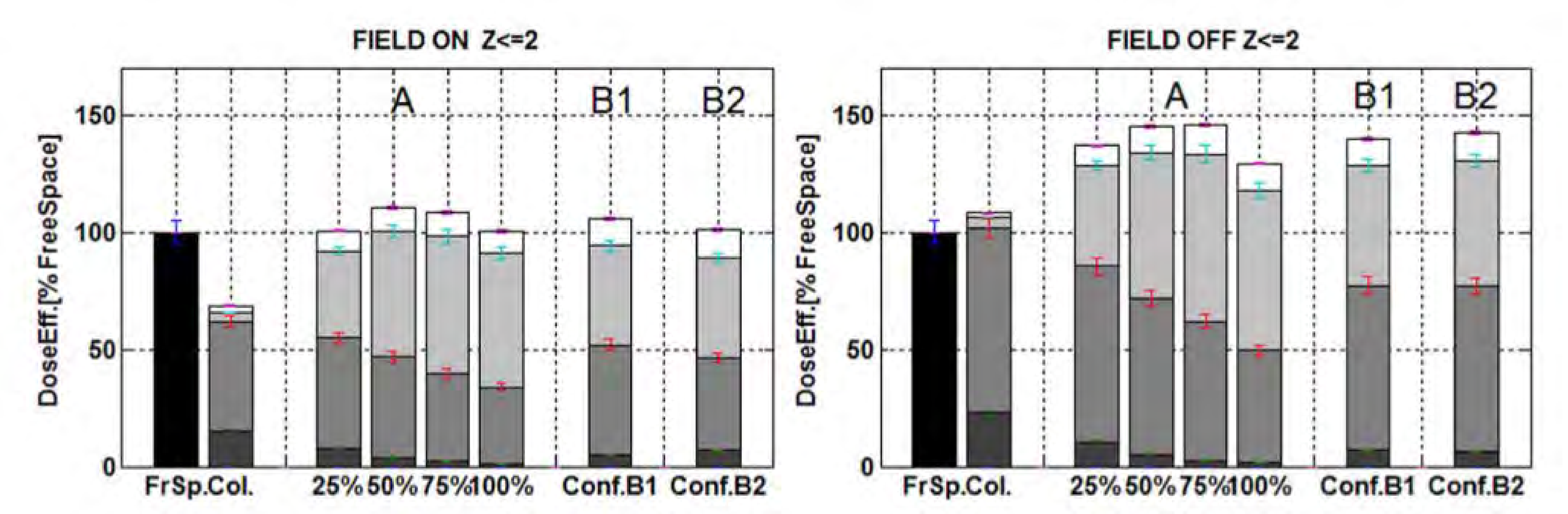


Bamford, R. A., et al. (2014)

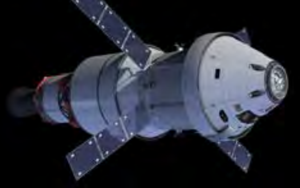
Drawbacks

- Thermal management of superconductors
- Danger of quench in proximity of habitat
- Hinders EVAs
- Re-designing Orion

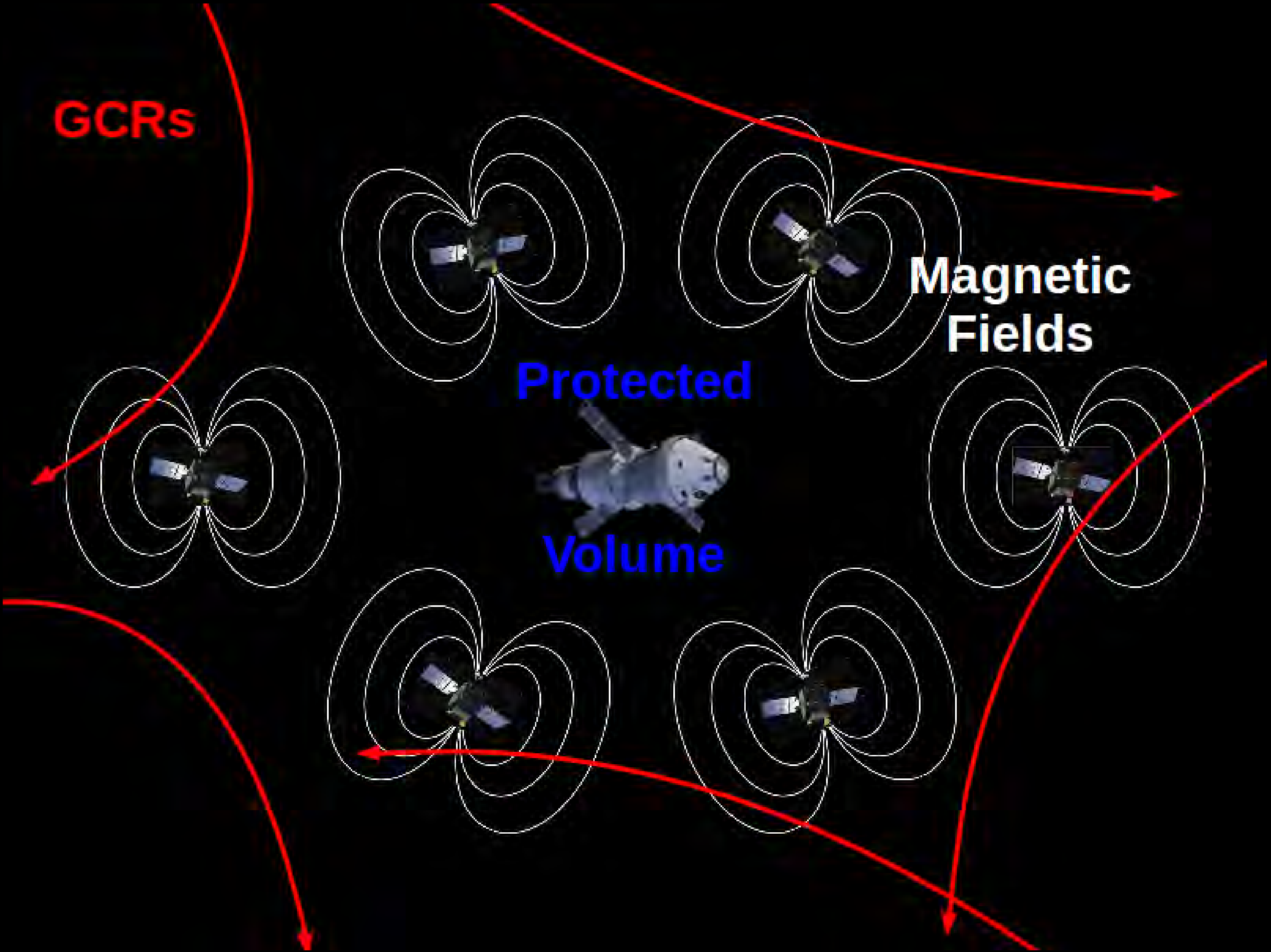
Increase of secondary radiation! (Vuolo et al. 2014)



New Concept



New Concept

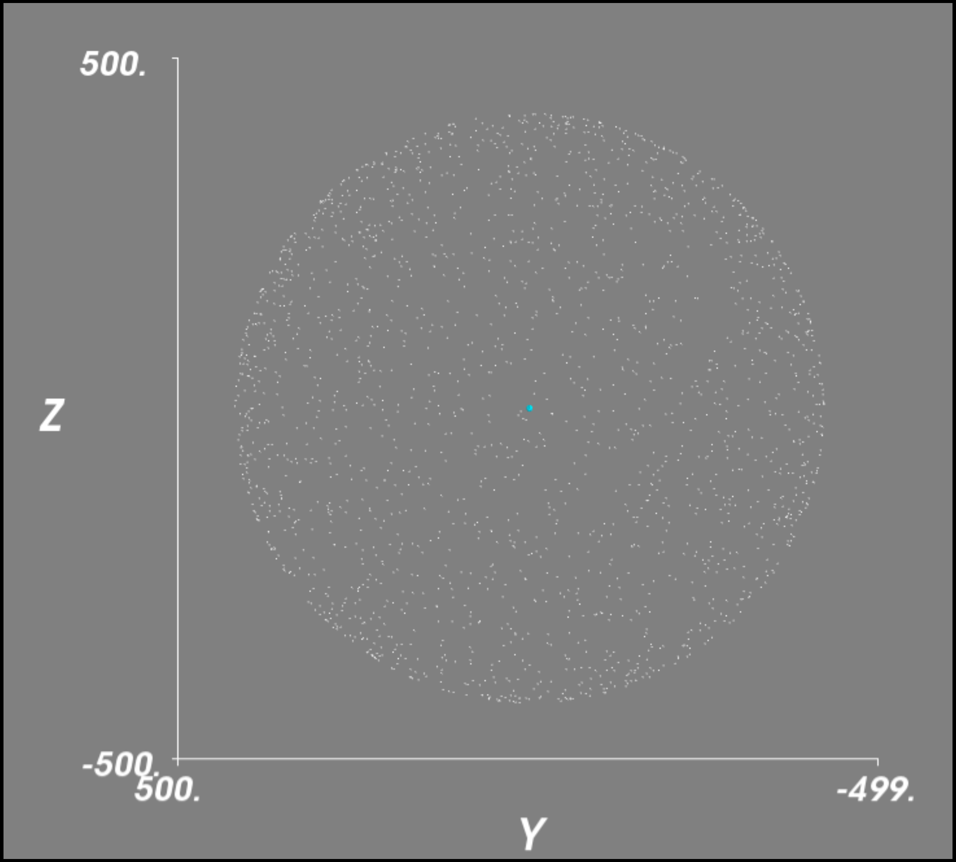


Goals

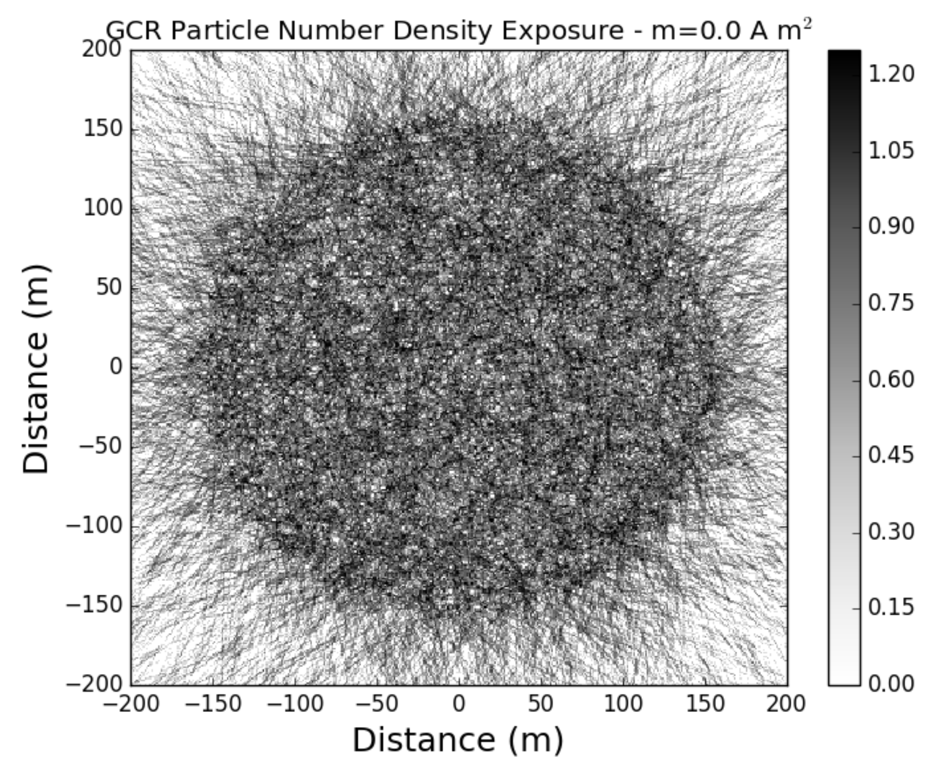
1. Optimize Design
2. First approximation
3. Good News
4. Great News
5. Bad News

Better solution?

Create Isotropic Environment



$$-1 < \cos\theta < 1$$
$$0 < \phi < 2\pi$$



Equation of Motion

$$\frac{d\vec{u}}{dt} = \frac{300}{E_n[\mathrm{MeV}]}\frac{Z}{A}(\vec{u} \times \vec{B}[T])$$

Energy space

Particle Advancement

$$\left.\begin{aligned} u[n+1] &= u[n] + a \cdot dt \\ r[n+1] &= r[n] + u[n+1] \cdot dt \end{aligned}\right\}$$

4th order
Runge Kutta

Form of the Magnetic Shield

$$B_x = \frac{Cxz}{2\alpha^2\beta\rho^2}\left[(a^2+r^2)E(k^2) - \alpha^2 K(k^2)\right]$$

$$B_y = \frac{Cyz}{2\alpha^2\beta\rho^2}\left[(a^2+r^2)E(k^2) - \alpha^2 K(k^2)\right]$$

$$B_z = \frac{C}{2\alpha^2\beta}\left[(a^2-r^2)E(k^2) + \alpha^2 K(k^2)\right]$$

CIRCULAR CURRENT LOOPS

Exact magnetic field solutions outside conductor

(Simpson et al. 2001)

magnetic moment $\mu = IA$

Remove particles that hit loops

$$r_{cs} < R^2 + a^2 - 2a(R^2 - z^2)^{1/2}$$

Energy loss

$$P = \frac{dE_n}{dt} = \frac{\mu_0 q^2 \gamma^6 u_0^2}{6\pi c}\left[c^4 a'^2 + |\vec{n} \times c^2 \vec{a}'|^2\right]$$

High Temperature Superconductors

YBCO

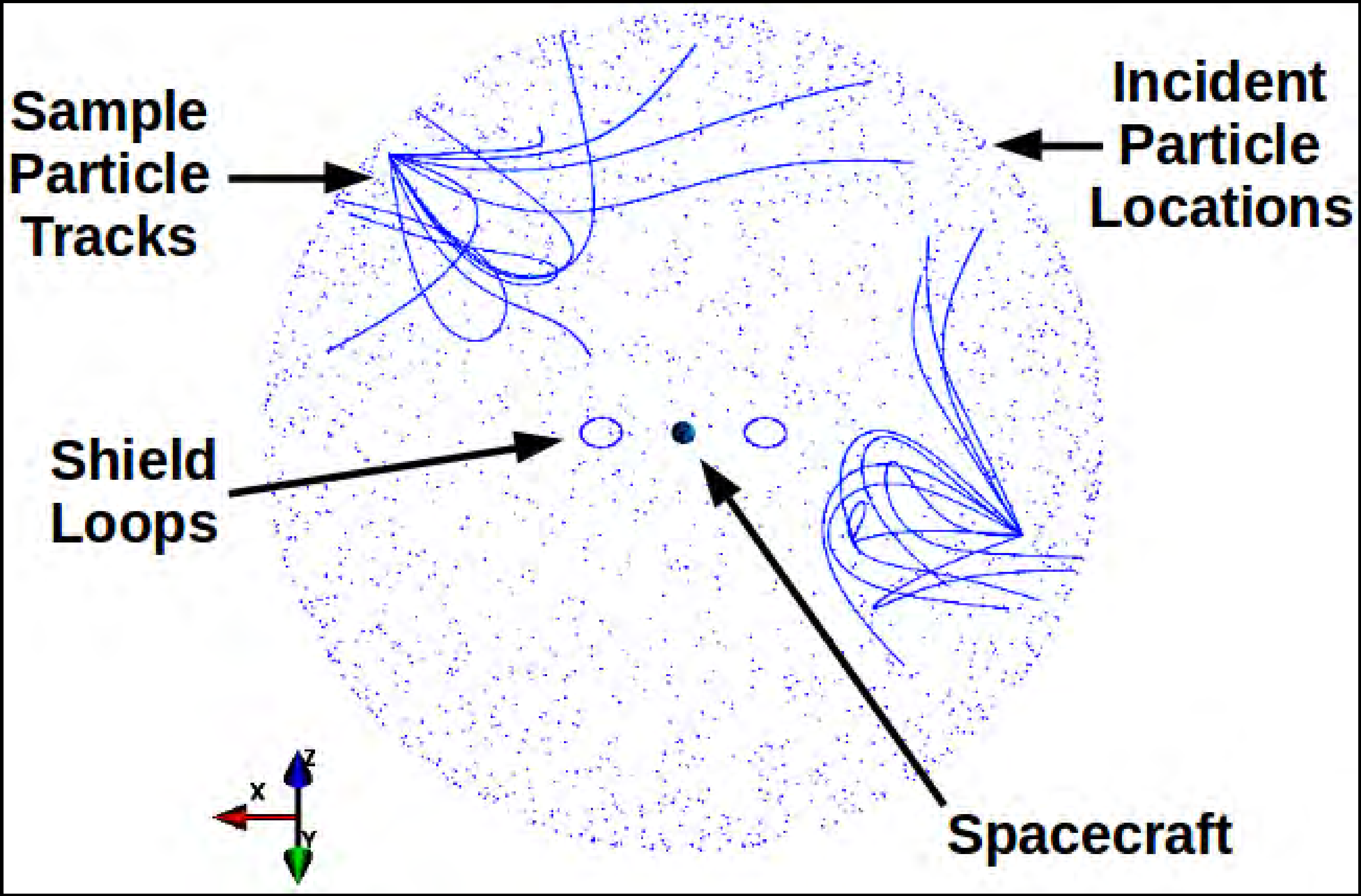
I = 300 A
J_e = 50 kA/cm^2
T_c = 90 K
T_o = 40-50 K

Loop dimensions

$$r_{cs} = \sqrt{\frac{\mu}{\pi^2 a^2 J_e}}$$

$$m = \frac{2\mu\rho}{aJ_e}$$

Configuration



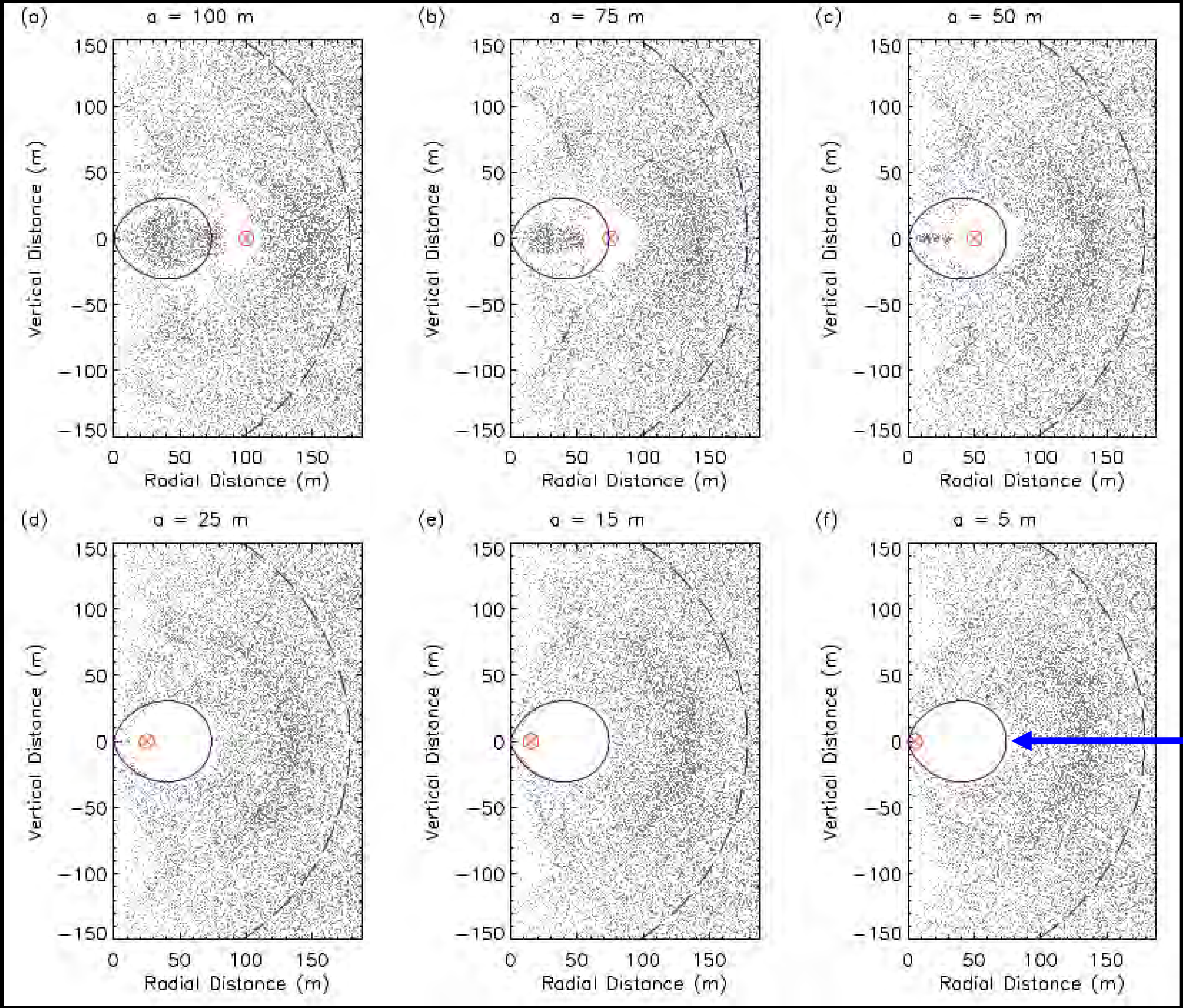
Sample Particle Tracks

Incident Particle Locations

Shield Loops

Spacecraft

Dispersed Shield – Large Loops



Shepherd & Kress (2007)

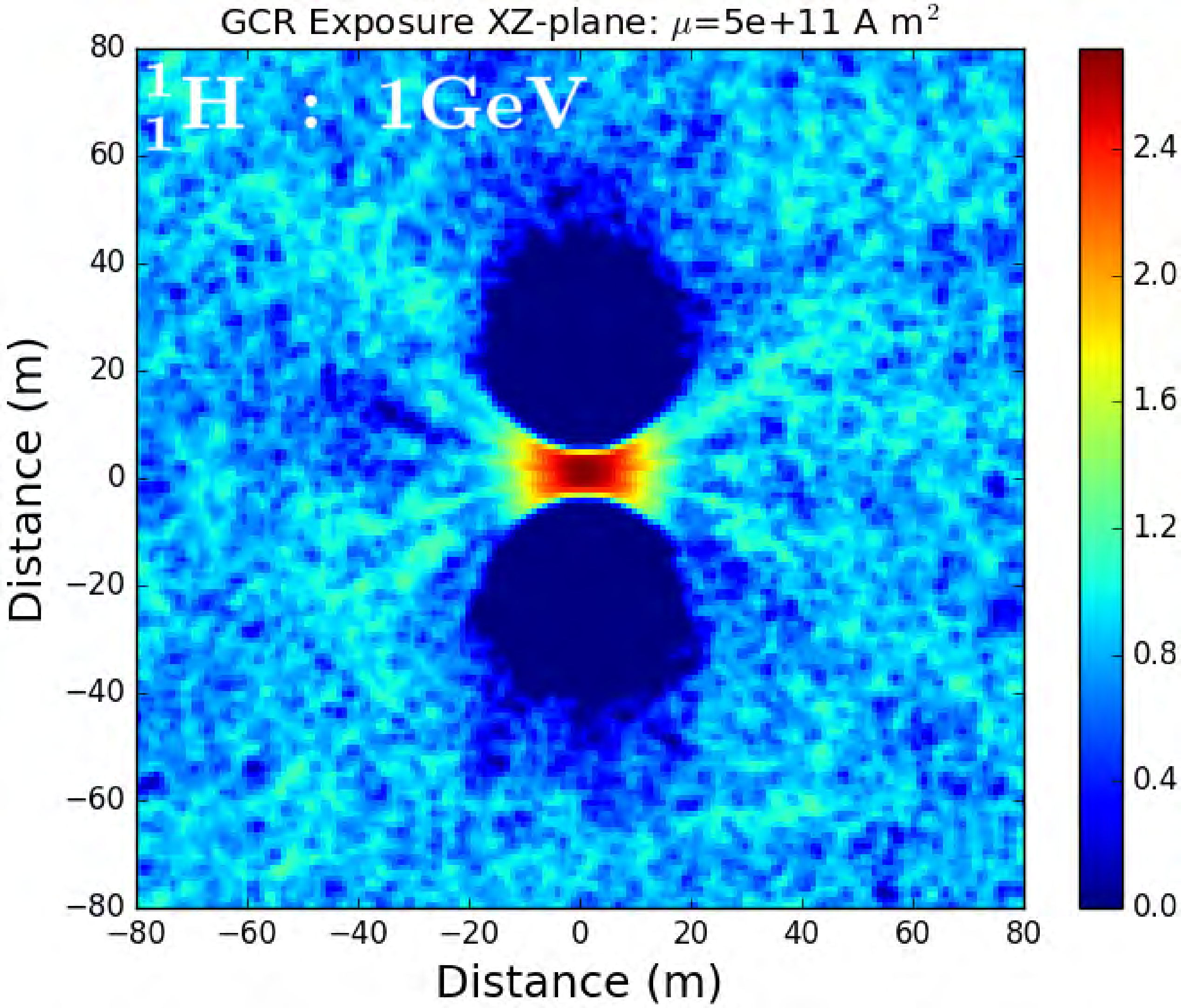
$\mu = 1.1\text{x}10^{13}$ A m^2

“Forbidden Zone”

Single Loop Simulations

1 GeV protons

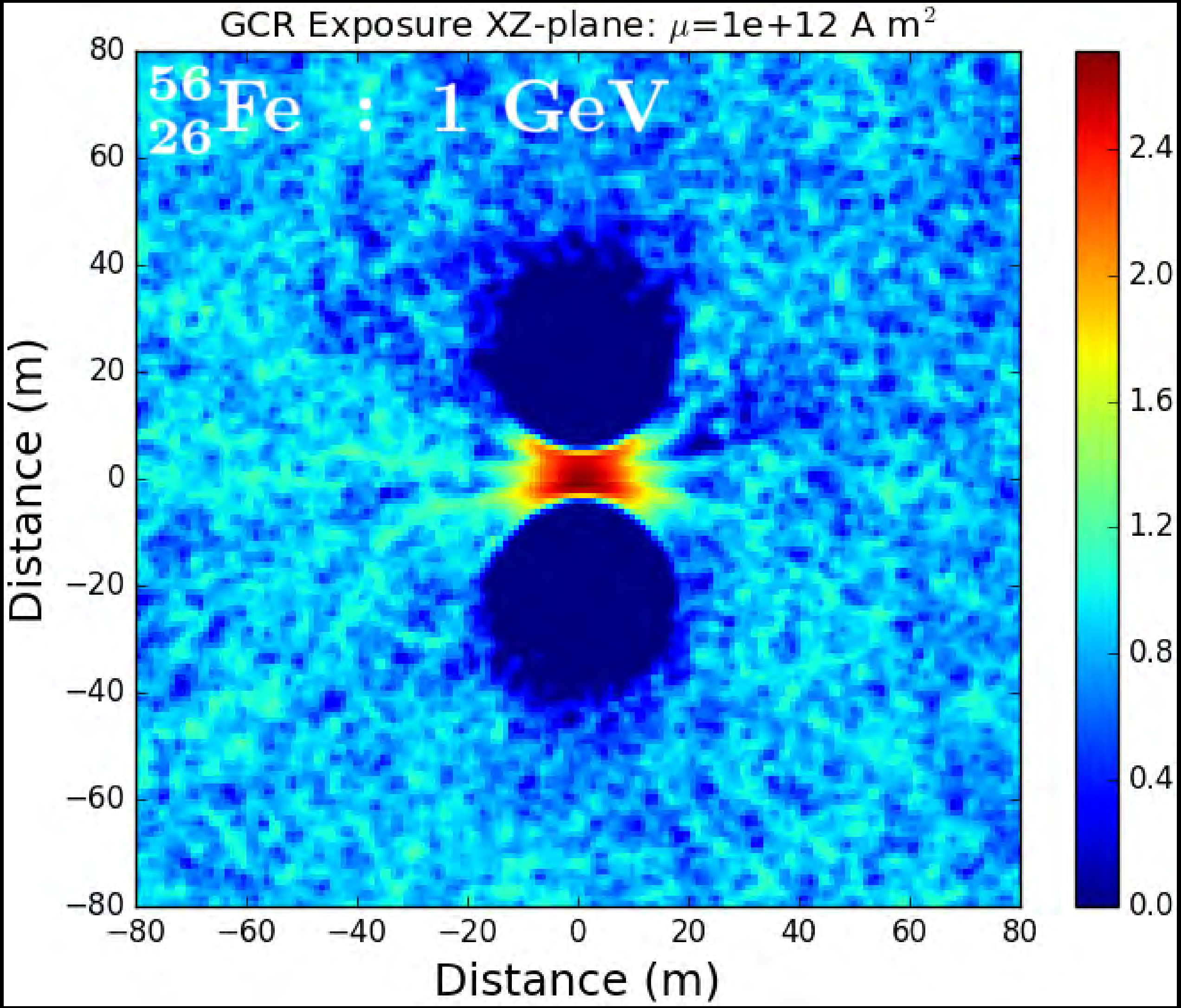
***a* = 10 m**



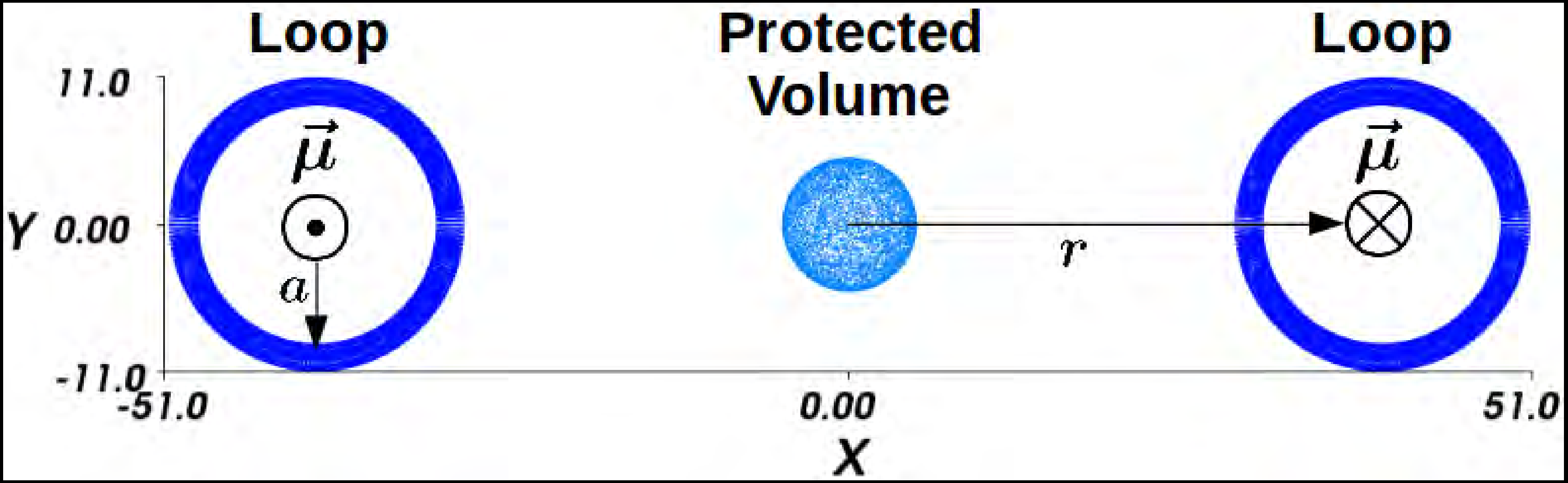
Single Loop Simulations

1 GeV
iron nuclei

***a* = 10 m**

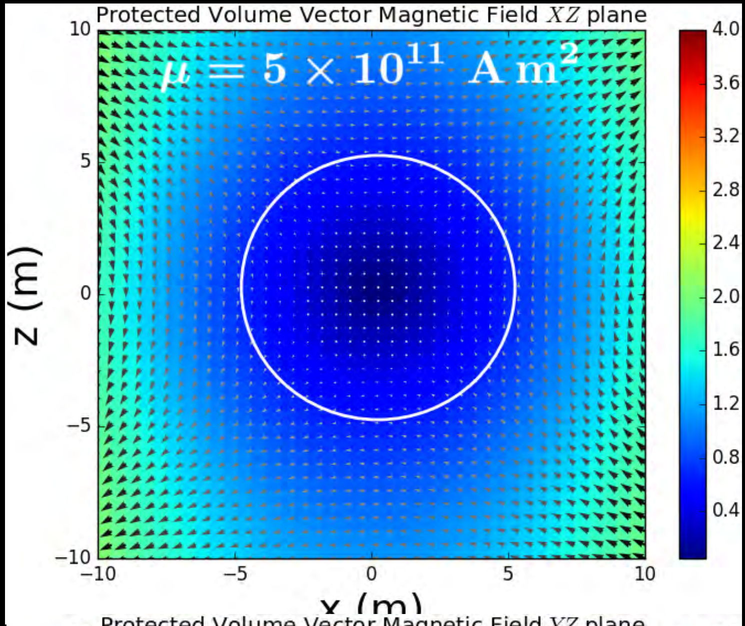


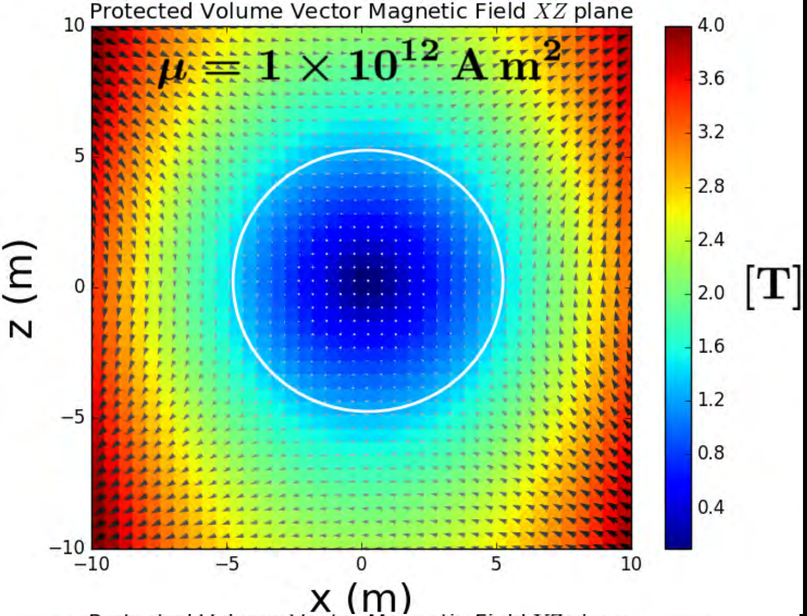
First Approximation



Two-loop magnetic “null”

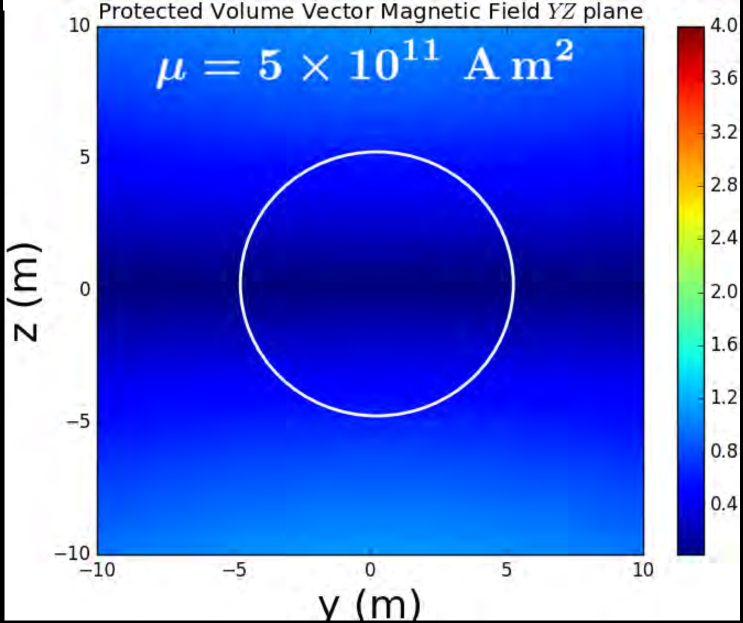
Magnetic Field Environment

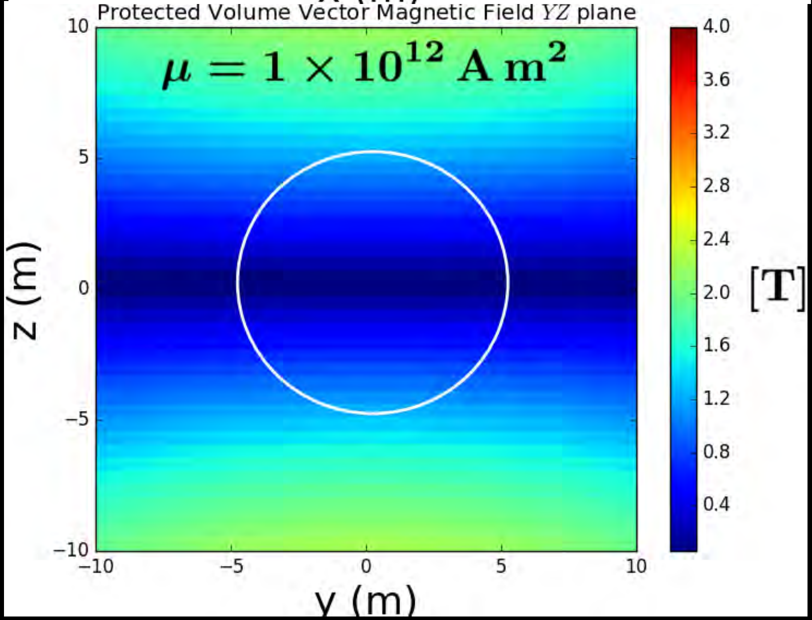




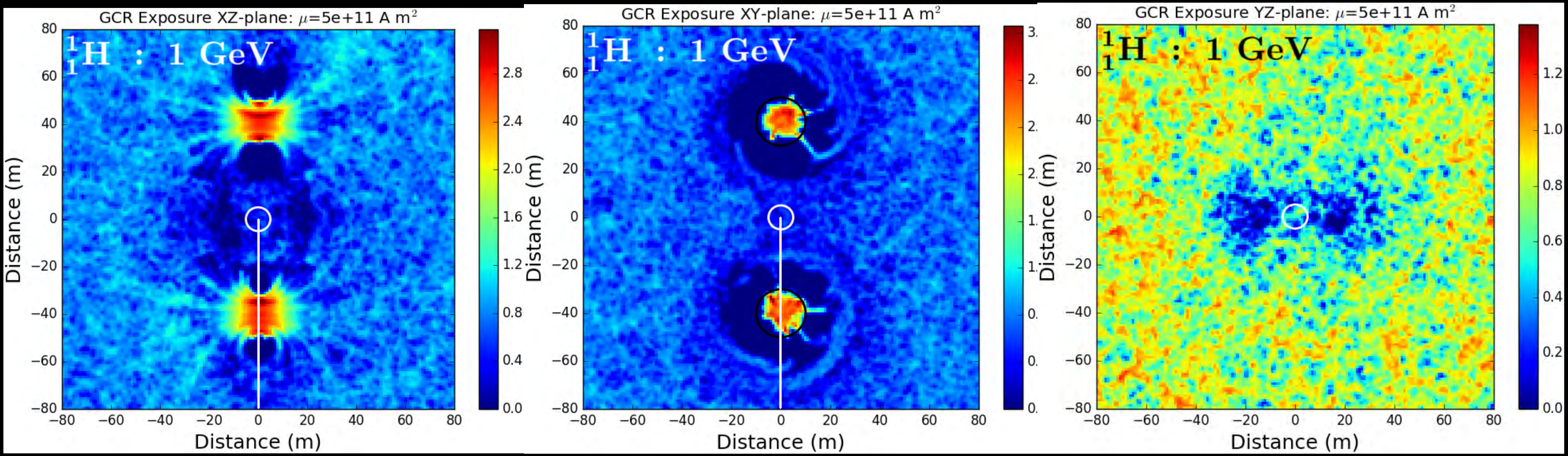
Deflect protons

Deflect iron nuclei



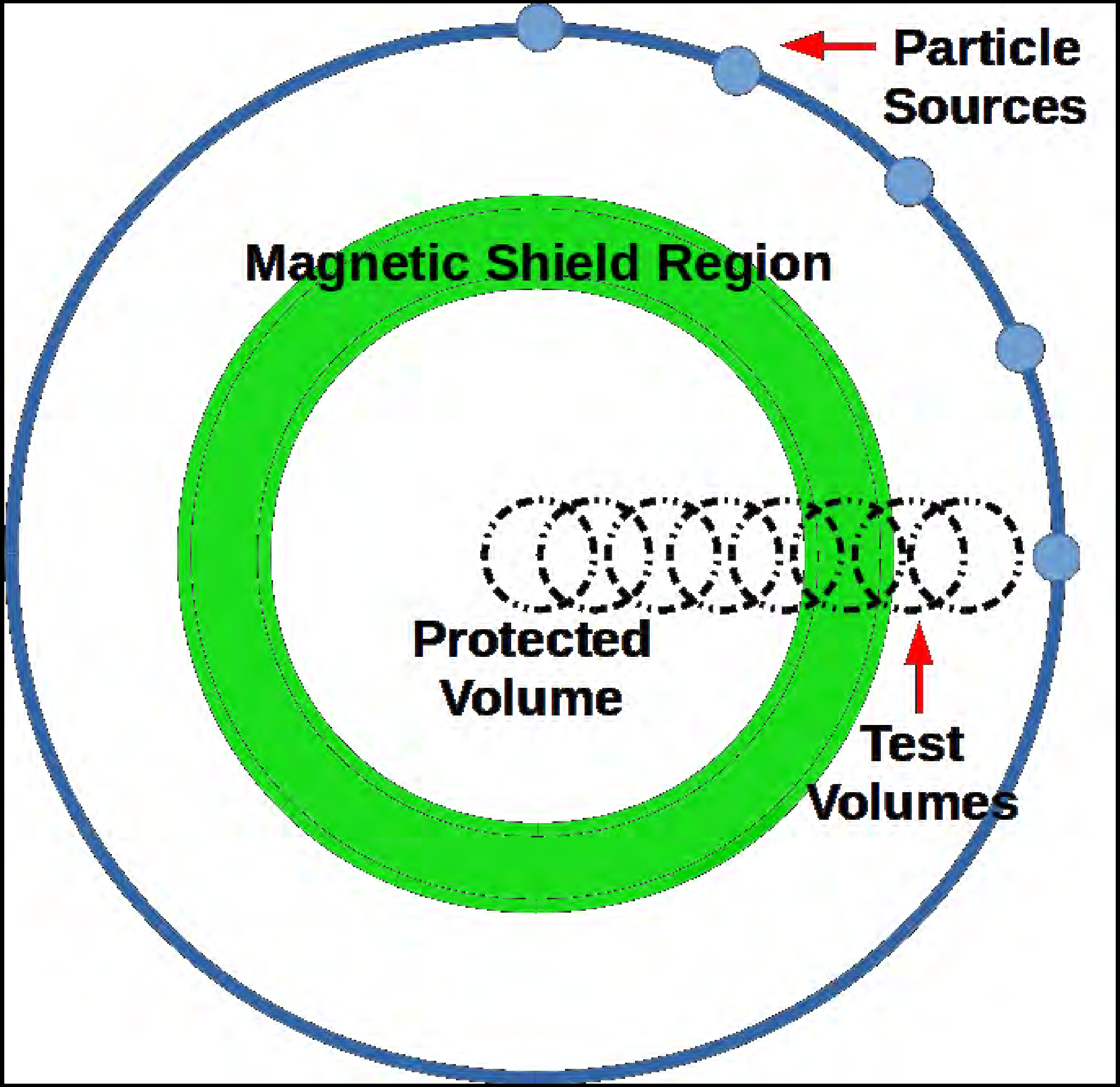


Simulations



Good news!

Diagnostics

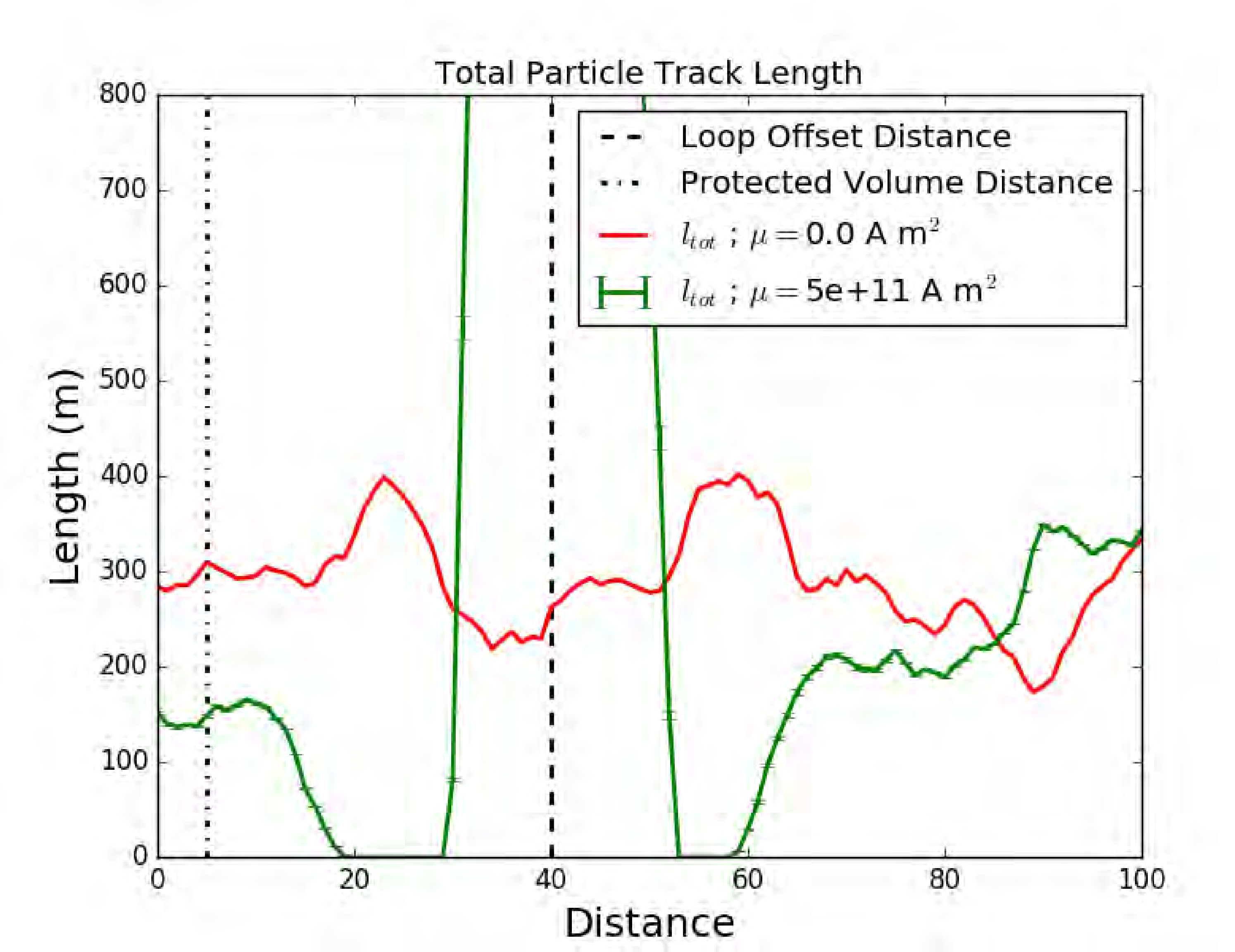


<u>**Track**</u>

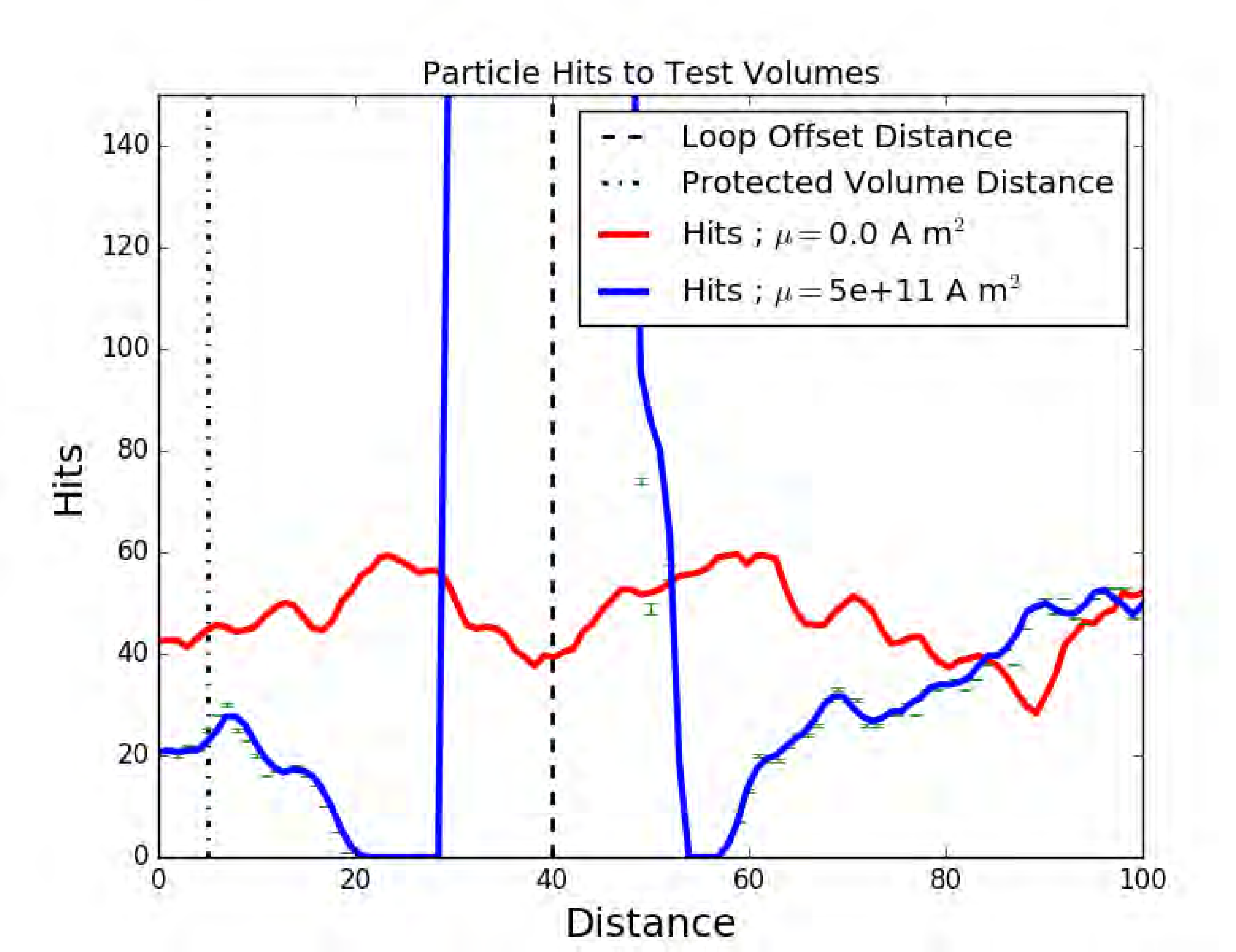
- Number of entering particles

- Total track length

Diagnostics

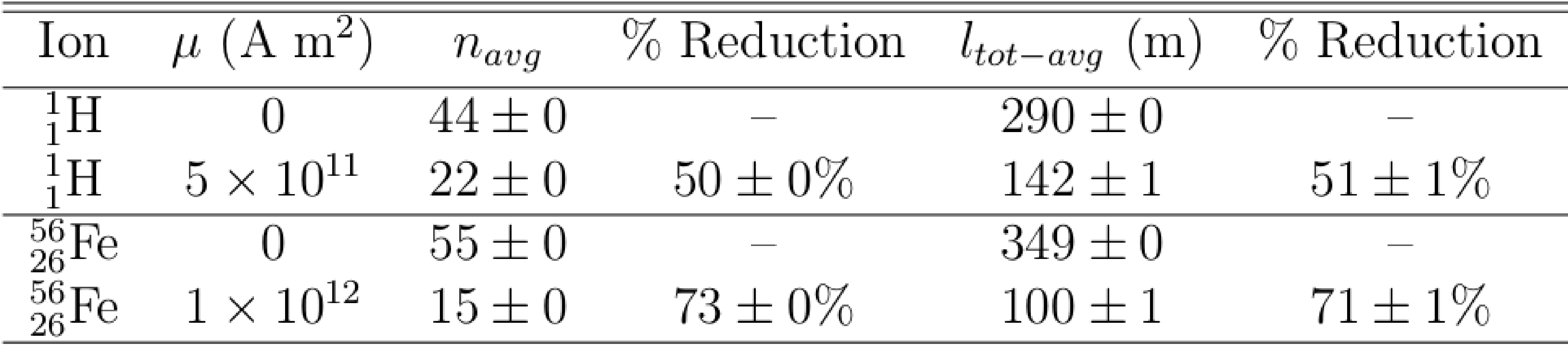


Diagnostics



Diagnostics

Protected Volume

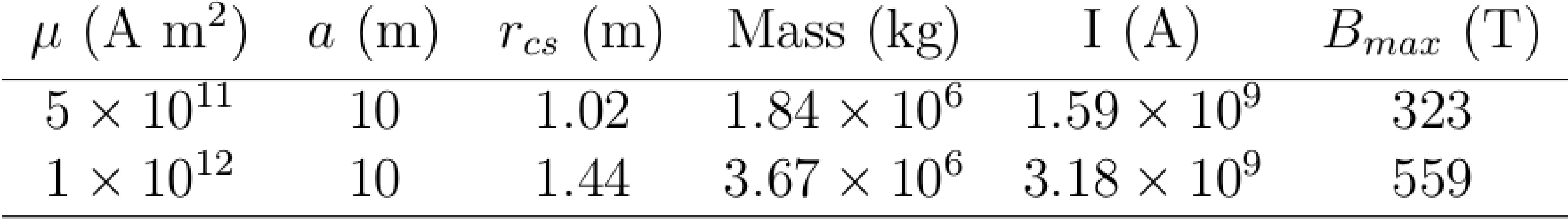


Ion	μ (A m^2)	n_{avg}	% Reduction	$l_{tot-avg}$ (m)	% Reduction
$^{1}_{1}$H	0	44 ± 0	–	290 ± 0	–
$^{1}_{1}$H	5×10^{11}	22 ± 0	$50 \pm 0\%$	142 ± 1	$51 \pm 1\%$
$^{56}_{26}$Fe	0	55 ± 0	–	349 ± 0	–
$^{56}_{26}$Fe	1×10^{12}	15 ± 0	$73 \pm 0\%$	100 ± 1	$71 \pm 1\%$

Great news!

Superconducting Loop Properties

Simulation Loops

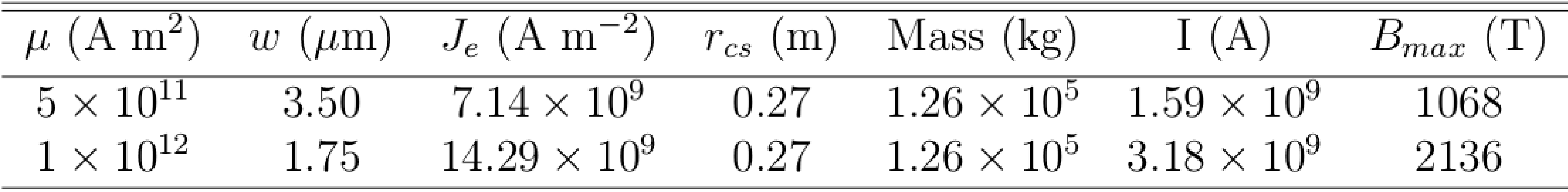


μ (A m^2)	a (m)	r_{cs} (m)	Mass (kg)	I (A)	B_{max} (T)
5×10^{11}	10	1.02	1.84×10^6	1.59×10^9	323
1×10^{12}	10	1.44	3.67×10^6	3.18×10^9	559

Bad news :(

NASA SLS Block 2 Payload = 130,000 kg

Alternative Loop Properties



μ (A m^2)	w (μm)	J_e (A m^{-2})	r_{cs} (m)	Mass (kg)	I (A)	B_{max} (T)
5×10^{11}	3.50	7.14×10^9	0.27	1.26×10^5	1.59×10^9	1068
1×10^{12}	1.75	14.29×10^9	0.27	1.26×10^5	3.18×10^9	2136

Optimization

1. “Decentralize” magnetic energy

- *more* dispersed = more loops

2. Reduce overall loop *I*, *B*, mass

3. Maximize the use of “forbidden zones”

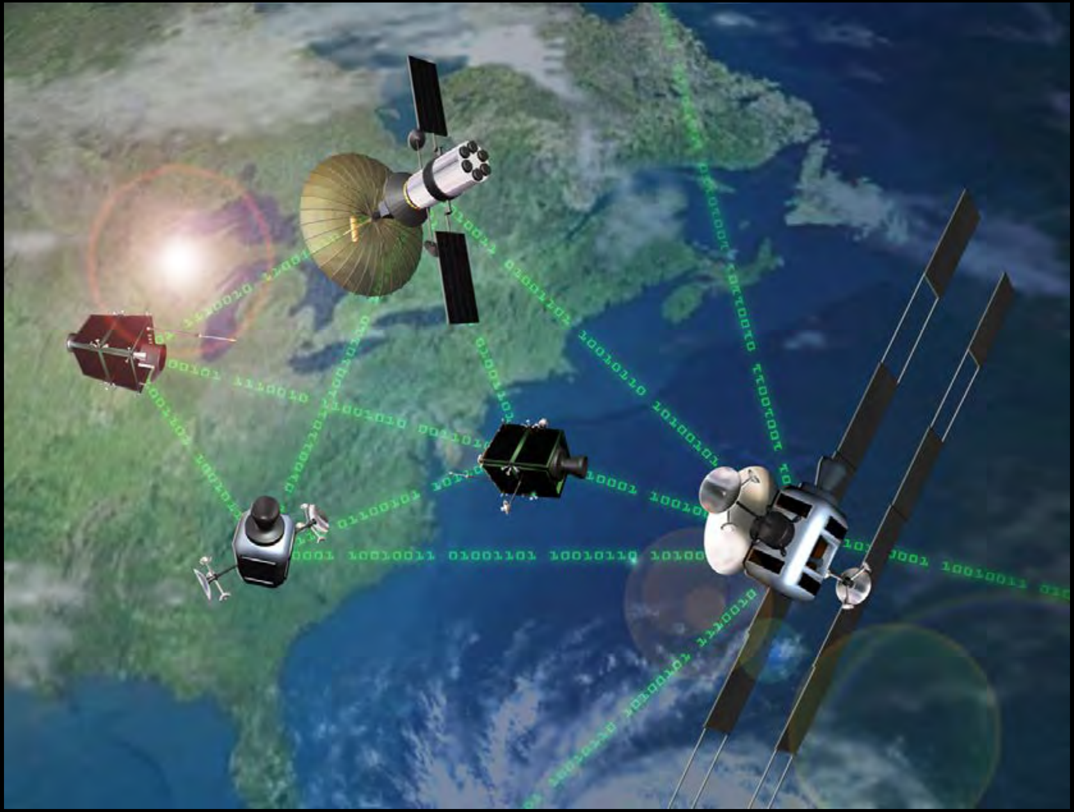
Further use of superconductors?

Superconductors in Space

- Superconducting magnetic energy storage (SMES)

- Docking and stability (magnetic levitation)

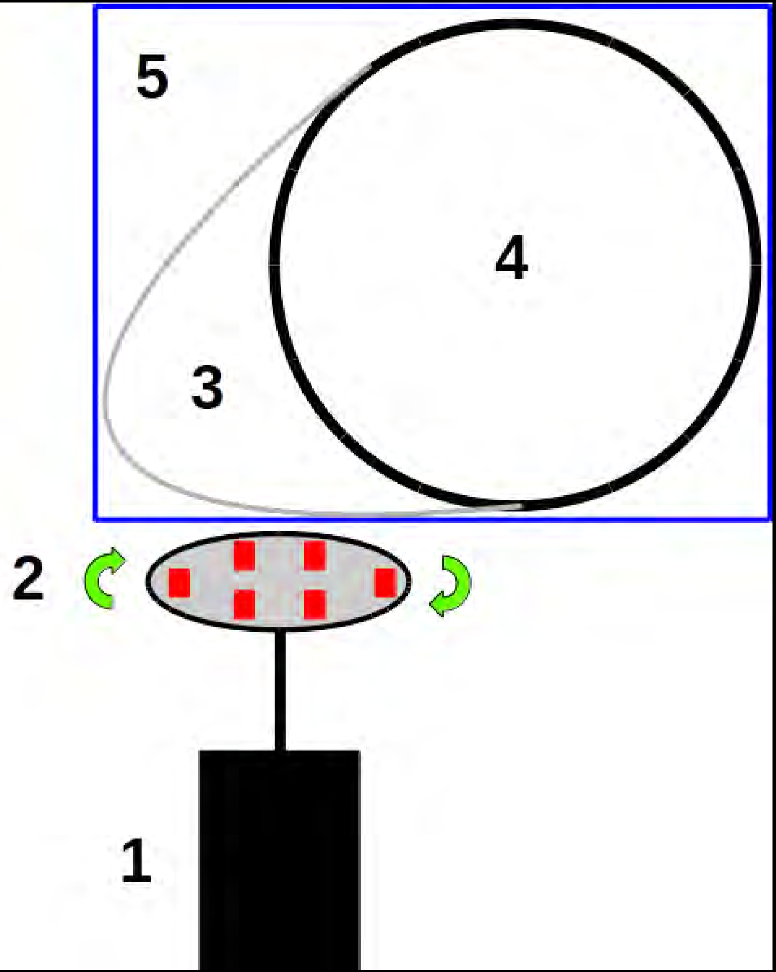
- Motors and MRIs



Charging Superconductors

Flux pumps

1. DC Motor
2. Rotor with Permanent Magnets (Red)
3. HTS Stator Loop With Soldered Joints
4. HTS Main Loop
5. Thermally Isolated Assembly



Conclusions

- Radiation mitigation is required for long duration exploration of space by humans

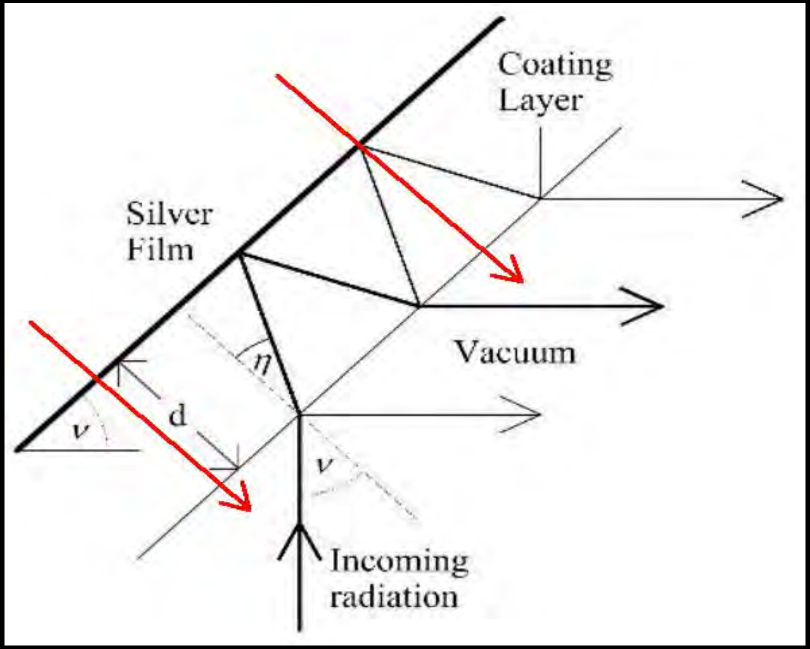
- Dispersed magnetic shield concept *works*, but needs *optimization*

- Synergistic combination of material shielding, magnetic shielding, and efficient propulsion

Thank You

orangewavedc@gmail.com

Thermodynamics



Youngquist & Nurge (2016)

$$J = \sigma A(T^4 - T_0^4)$$

$$\Delta E = C\Delta T + \sigma A(T^4 - T_0^4)\Delta t$$

Attenuation coefficient $\alpha = \dfrac{J}{J_0}$

$$J = \frac{\Delta E}{\Delta t}$$

Limits of passive cooling

Cryogenic Select Surfaces

Reflects 99.9% solar irradiance

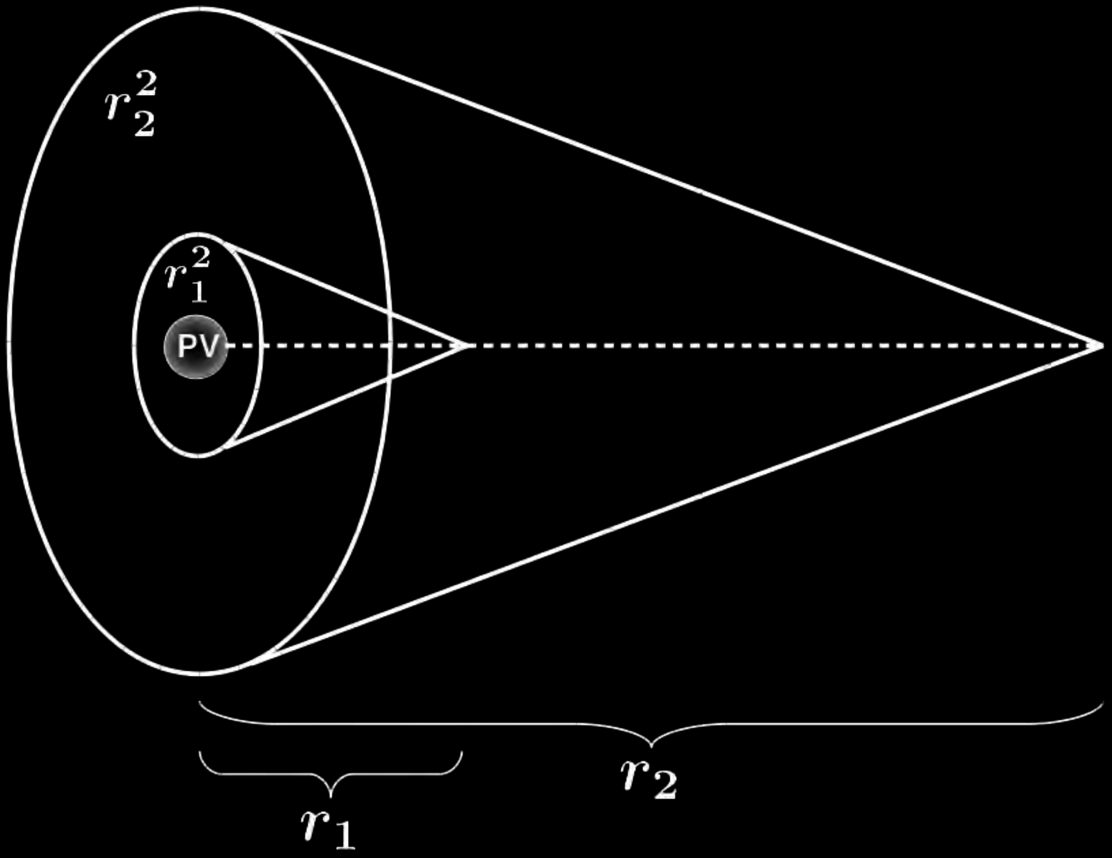
Transmits long infrared radiation from interior

Result: Cryogenic temperatures below 50K

Secondary Particle Threat

$$\Omega_{PV} = \oint F_{\Omega} d\phi$$

$$\Omega_{PV} = \frac{A_{PV}}{r_{off}^2} \oint d\phi$$



Other Solution?

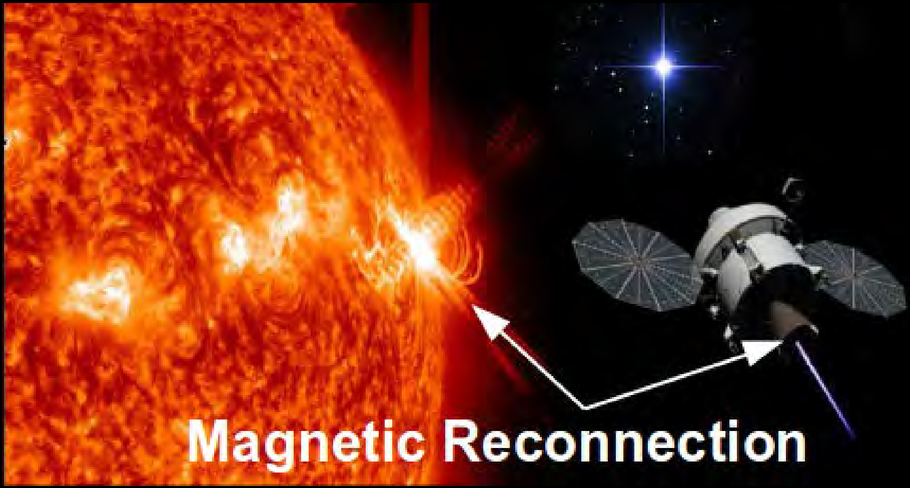
In the absence of a radiation shield – reduce exposure *time*

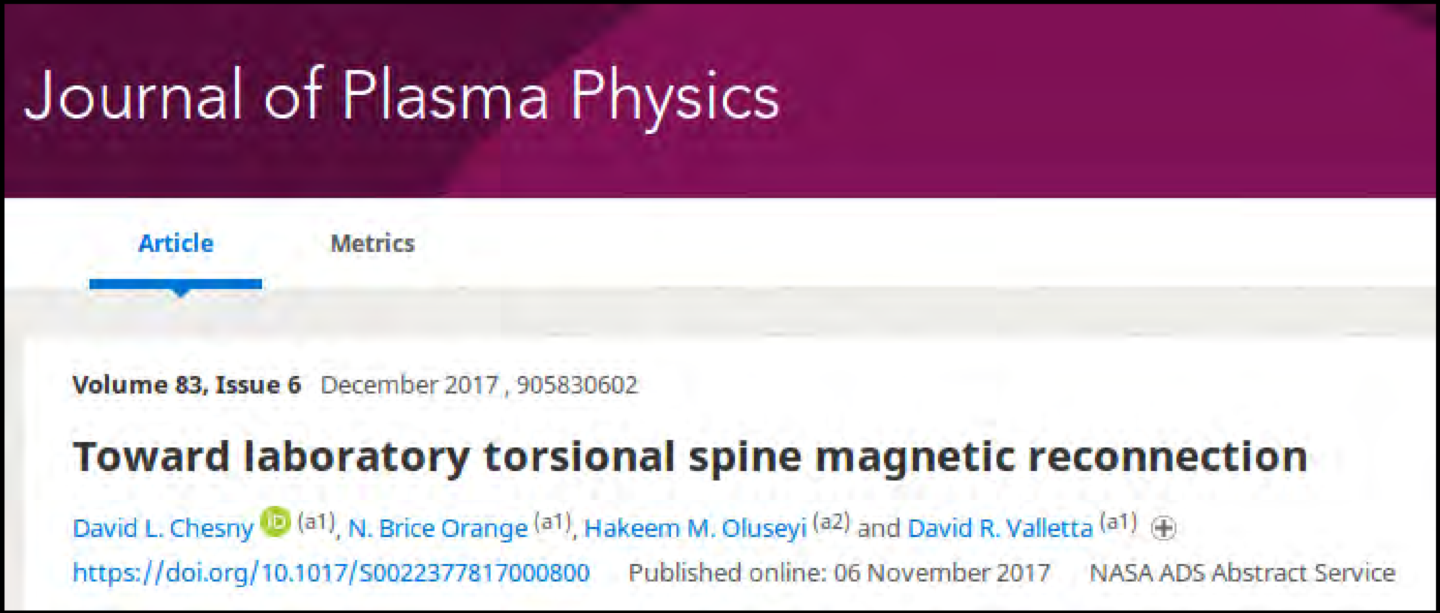
Increase thrust!

$$T = \dot{m} v_{ex}$$

What is the most efficient particle acceleration process in the solar system?

Magnetic Reconnection




Journal of Plasma Physics

Volume 83, Issue 6 December 2017 , 905830602

Toward laboratory torsional spine magnetic reconnection

David L. Chesny (a1), N. Brice Orange (a1), Hakeem M. Oluseyi (a2) and David R. Valletta (a1)

https://doi.org/10.1017/S0022377817000800 Published online: 06 November 2017 NASA ADS Abstract Service