

G.U. Aero Report 9214

#### Store

# Engineering PERIODICALS

# **DEPARTMENT OF AEROSPACE ENGINEERING**

UNIVERSITY OF GLAS

# COLLECTED DATA FOR TESTS ON AN AHAVAW AEROFOIL

**VOLUME III :** Pressure data relevant to the study of large-scale vertical-axis wind-turbines.

by

# JIANG DACHUN R.A.McD.GALBRAITH F.N.COTON M.W.GRACEY and

ina

**R.GILMOUR** 

March 1992

# COLLECTED DATA FOR TESTS ON AN AHAVAW AEROFOIL

Herein is presented the collected data for tests in which an AHAVAW aerofoil was subjected to a variety of oscillatory displacements in pitch about the quarter-chord location at low Reynolds numbers.

### VOLUME III

# PRESSURE DATA RELEVANT TO THE STUDY OF LARGE-SCALE VERTICAL-AXIS WIND-TURBINES.

by

#### JIANG DACHUN

#### R.A.McD.GALBRAITH

#### F.N.COTON

#### M.W.GRACEY

and

#### R.GILMOUR

March 1992

# AHAVAW - VOLUME III

# CONTENTS

|   | Nomenclature                           | • •   | •   | • • | ٠  | • | ••• | • | 1             |
|---|--|-------|-----|-----|----|---|-----|---|---------------|
| 1 | 1 Introduction                         | •     | •   | • • | •  | • |     | • | 1             |
| 2 | 2 Description of Test Facility         |       |     |     | •  | • |     | • | 1             |
|   | 2.1 Aerofoil and Wind Tunnel           |       |     |     |    |   |     | • | 1             |
|   | 2.2 Pitch-drive Mechanism              |       |     |     | •  |   |     |   | 1             |
|   | 2.2.1 Actuator                         |       | •   |     |    |   |     |   | 1             |
|   | 2.2.2 Command Signal                   | • •   | •   | • • |    | · | · · | · | 2             |
|   | 2.3 Instrumentation and Data Logging . | • •   | ·   | • • | •  | • | • • | • | 2             |
|   | 2.3.1 Pressure Transducers             | · · · | · • | • • | ·  | • | • • | 2 | $\frac{2}{2}$ |
|   | 2.3.3 Incidence                        |       | :   |     | ÷  | • | · · |   | $\frac{2}{2}$ |
|   | 2.3.4 Acquisition Unit                 |       | •   |     |    | • |     |   | 2             |
|   | 그 일을 물고 있는 것이 같이 많이 많이 많이 했다.          |       |     |     |    |   |     |   |               |
| 3 | 3 Test Series and Procedure            | •     | •   | • • | •  | • | • • | • | 3             |
|   | 3.1 Static Experiment                  | •     | •   |     |    |   |     | • | 3             |
|   | 3.2 Sinusoidal Experiment              |       |     |     | •  | • |     |   | 3             |
|   | 3.3 VAWT Experiment                    |       |     |     | ÷. |   |     |   | 3             |
|   | 3.4 Procedure                          |       | •   |     |    |   |     |   | 4             |
|   | 3.5 Roughness Transition Strips        | •     | ÷.  |     |    |   |     |   | 4             |
|   | 3.6 Data Presentation                  |       |     |     | •  |   |     |   | 4             |
|   |  |       |     |     |    |   |     |   |               |
| 4 | 4 Results and Discussion               | •     | •   |     |    | • |     |   | 4             |
|   | 4.1 Tunnel Performance                 |       |     |     |    |   |     |   | 4             |
|   | 4.2 Averaging of the Data              | •     |     |     |    |   |     |   | 5             |
|   | 4.3 Test Data                          |       |     |     |    |   |     |   | 5             |
|   |  |       |     |     |    |   |     |   |               |
| A | Acknowledgements                       | •     |     |     |    |   |     |   | 5             |
|   |  |       |     |     |    |   |     |   |               |
| R | References                             |       |     |     |    |   |     |   | 5             |
|   |  |       |     |     |    |   |     |   |               |
| Т | Tables                                 |       | •   |     | •  |   |     |   | 6             |
|   |  |       |     |     |    |   |     |   |               |

Figures

#### NOMENCLATURE

| с    | chord                               |
|------|-------------------------------------|
| Cm   | pitching-moment coefficient         |
| Cn   | normal force coefficient            |
| Cp   | pressure coefficient                |
| Ct   | "thrust" force coefficient          |
| D.P. | dynamic pressure ( $\rho V^2/2$ )   |
| k    | reduced frequency ( $\omega c/2V$ ) |
| r    | reduced pitch-rate (c/2V)do/dt      |
| TSR  | tip speed ratio                     |
| Re   | Reynolds number                     |
| V    | velocity                            |
| x/c  | chordwise dimension                 |
| α    | angle of attack                     |
| ω    | rotational velocity                 |

#### **1 INTRODUCTION**

At present, in the United Kingdom, United States of America and Canada, vertical-axis wind turbines (VAWTs) typically employ the NACA 0015 aerofoil for the turbine rotors. If thicker sections could be shown to be aerodynamically satisfactory, their use would lead to a simplification in the blade design and, hence, cost reductions. As a result, the University of Glasgow is currently researching the effect on aerodynamic characteristics of varying aerofoil thickness.

As part of this investigation, in the dynamic stall facility at the University of Glasgow<sup>1,2,3</sup>. two-dimensional data has been acquired from experiments on a number of aerofoils under a variety of motion types. Angell et al<sup>4</sup> obtained relevant lift, thrust and pitchingmoment data for five aerofoil sections (NACA 0015, NACA 0018, NACA 0021, NACA 0025 and NACA 0030). From analysis of data produced by experiments on these aerofoils, a second generation of aerofoil sections have been designed. This report, the third of three, presents the collected data from a series of oscillatory tests performed on a new aerofoil, designed by Vertical Axis Wind Turbines Limited, which is a member of the second group. The coordinates for this aerofoil section, named the AHAVAW, are listed in The experiments are split between Table 1. the three volumes as follows:

**VOLUME I** Pressure data from ramp function tests.

**VOLUME II** Pressure data from oscillatory tests.

**VOLUME III** Pressure data relevant to the study of large-scale vertical-axis wind turbines.

Each volume also includes the pressure data from tests in steady conditions and a brief description of the experimental apparatus and techniques.

#### 2 DESCRIPTION OF TEST FACILITY

#### 2.1 Aerofoil and Wind Tunnel

The general arrangement of the aerofoil in the wind tunnel was as shown in **Figure 1**. The aerofoil, of chord length 0.55m and span 1.61m, was constructed of fibre glass mounted on an aluminium spar and filled with an epoxy resin foam. The hand-finished surface was very smooth, and the profile accurate to better than 0.1mm. The instrumented model was fitted vertically into the University of Glasgow's "Handley Page" wind tunnel.

The "Handley Page" low-speed wind tunnel is an atmospheric-pressure closed-return type with a 1.61x2.13 octagonal working section (**Figure 2**) in which a wind velocity of 61ms<sup>-1</sup> can be attained. The model was pivoted about its quarter-chord axis on two tubular steel shafts connected to the main support via two selfaligning bearings. A single thrust bearing on the top support beam took all the weight. The dynamic and aerodynamic loadings from the aerofoil were reacted to the tunnel framework by two transversely mounted beams.

#### 2.2 Pitch Drive Mechanism

#### 2.2.1 Actuator

Angular movement of the model was obtained using a linear hydraulic actuator and crank mechanism. The actuator was mounted horizontally below the tunnel working section on the supporting structure, with the crank rigidly connected to the tubular part of the spar by a welded sleeve and keyway. The acuator was a UNIDYNE 907/1 type with a normal dynamc thrust of 6.1KN operated from a supply pressure of 7.0MNm<sup>-2</sup>. A MOOG 76 series 450 servo valve was used via a UNIDYNE servo controller unit to control the movement of the actuator. A suitable feedback signal for the controller was provided by a precision linear angular displacement transducer geared to the main spar of the model.

#### 2.2.2 Command Signal

The model's angle of attack was incremented by the actuator controller. The input signal during the static tests was provided under software control by the data acquisition unit's own digital-to-analogue converter. This was possible because, during the sampling, the angle of attack was fixed and sufficient time was available between sampling to set the model at the required angle of attack. The two activities were separate and were performed sequentially.

Such was not the case during the unsteady tests, however, where sampling and control of were required motion model's the simultaneously. Therefore, during these tests, the input signal was provided by a separate function generator, comprised of an AMSTRAD 1512 microcomputer equipped with an ANALOG DEVICES RTI815 multi-function The required output input/output board. function was digitised into equal time steps in 2's complement code and the frequency of the function was controlled using the internal interrupts of the AMSTRAD microcomputer. The code was written in TURBO PASCAL.

#### 2.3 Instrumentation and Data Logging

#### 2.3.1 Pressure Transducers

To provide the chordwise pressure distribution at mid-span, thirty KULITE XCS-093-5 PSI G ultra-miniature pressure transducers were installed just below the surface of the centre section of the model. The transducers were of vented gauge type with one side of the pressure sensitive diaphragm open to the ambient pressure outside the wind-tunnel (via tubes in the model). Each transducer was fitted with a temperature compensation module, which minimised the change in zero-offset and sensitivity with temperature. The locations of the pressure transducers in the model are illustrated in Figure 3.

The low voltage outputs from the thirty presure transducers were suitably amplified and conditioned by a bank of differential amplifiers. The conditioned signals were passed to a "sample and hold" unit<sup>1,5</sup> to overcome the timeskew problem arising from the sequential conversion of the anlogue signals into digital form.

#### 2.3.2 Dynamic Pressure

The dynamic pressure in the wind tunnel working section was determined by measuring the difference between the static pressure in the working section, 1.2m upstream of the leading edge, and the static pressure in the settling The pressure tappings were chamber. FC012 FURNESS connected to a micromanometer, which provided an analogue signal suitable for the data acquisition unit's analogue-to-digital converter. This dynamic pressure was recorded as the sample-and-hold unit was triggered to sample the output from the pressure transducers.

#### 2.3.3 Incidence

The instantaneous angle of attack of the aerofoil was determined by an angular displacement transducer geared to the model's main spar. The signal voltage from the transducer was fed into an amplifier/splitter to produce three signals for the following purposes:

- i) connection of the multiplexer for recording the aerofoil's angle of attack;
- ii) connection of the Schmitt trigger for initiation of data sampling when a preset incidence (voltage) was attained;
- iii) a feedback signal to the hydraulic actuator controller.

#### 2.3.4 Acquisition Unit

The actual data acquisition unit was a DEC MINC-11 microcomputer, configured with an LSI-11/32 16-bit microprocessor and laboratory modules which included:

- i) an analogue-to-digital converter module, with a 16-channel multiplexer incorported. The converter was a 12-bit successive approximation type with a conversion time of of  $30\mu s$ , but the multiplexer's settling time and the need to transfer the data from the analogue-todigital converter into system memory increased the conversion time to  $44\mu s$ ;
- ii) a multiplexer module, of 16 single-ended channels, which increased the number of channels that could be sampled to 32;
- iii) a real-time clock module, with two Schmitt triggers. This was used as a time-base generator to accurately set the sampling frequency. The sampling frequency was determined at run time from the frequency of oscillation and the requirement that 128 sample sweeps should be obtained during each cycle. One of the Shmitt triggers was used to initiate data sampling, by setting its reference voltage to a value corresponding to the angular displacement transducer's output for the required mean angle of attack;
- iv) a digital-to-analogue converter module which housed four independent 12-bit digital to analogue converters. This was used to provide the command signal for the hydraulic actuator during static tests.

The path of data flow and system layout is shown diagrammatically in **Figure 4**. The main control programs for the tests were written in FORTRAN IV, as described by **Murray-Smith and Galbraith**<sup>6</sup>. The programs prompt the user for specific run information before calling a specialised subroutine written in MACRO-11 assembly language to receive and store the digitised data. The timing and control of the analogue-to-digital converter and associated circuitry was performed by the processor's hardware, but channel selection and data management were achieved under software control.

#### 3 TEST SERIES AND PROCEDURE

#### 3.1 Static Experiment

A number of experiments were performed under steady conditions. Once the wind velocity had reached the required value, the aerofoil was rotated about its guarter-chord axis until it was positioned at the incidence at which the first set of data were to be recorded. Usually, this was approximately  $-2^{\circ}$ . The model's angle of attack was then increased in steps of approximately 0.5°. After each increment in incidence, the flow was allowed to stabilise for a few seconds before each transducer's output was sampled 100 times and the mean value for each was stored. After 64 sweeps of data had been recorded, the model was returned to its starting position. Data sampling was maintained at the same rate on the return arc in order to record any delay in the reattachment of flow.

#### 3.2 Sinusoidal Experiment

For this experiment, the model was rotated about its quarter-chord axis so that its angle of attack varied sinusoidally with time. The amplitude and frequency were controlled by the AMSTRAD function generator. During each oscillatory cycle 128 data sweeps were recorded and logged, with data being sampled during ten cycles.

#### **3.3 VAWT Experiment**

The VAWT experiment was designed to emulate the incidence time histories encountered by the blade of a vertical-axis wind turbine. A computer algorithm, coded in FORTRAN 77, has been developed at the University of Reading to calculate the blade's angle of attack as a function of its azimuth position. The program can use both single and multiple streamtube models<sup>7</sup> based on SANDIA<sup>8</sup> data for the NACA series of aerofoil characteristics. At low tip-speed ratios the time history for the single streamtube model is a skewed sine function, but this tends toward a true sine as the tip-speed ratio is increased. The upwind (positive) and downwind (negative) sections of each cycle attain identical peak values of incidence. Tip-speed ratio and amplitude are related as follows:

| TSR  | Amplitude         |
|------|-------------------|
| 6.00 | 5.4 <sup>0</sup>  |
| 4.00 | 9.9 <sup>0</sup>  |
| 3.50 | 12.2 <sup>0</sup> |
| 3.25 | 13.8 <sup>0</sup> |
| 2.80 | 17.4 <sup>0</sup> |
| 2.33 | 22.6 <sup>0</sup> |
| 1.75 | 32.8 <sup>0</sup> |

The AMSTRAD function generator reproduced the angle of attack histories based upon the NACA 0015 aerofoil's characteristics. Data acquisition was performed in an identical manner to that for sinusoidal tests.

In addition, a number of non-standard VAWT experiments were performed. Each is described in **Table 5**.

#### 3.4 Procedure

Before each individual set of tests, the tunnel was shut down and the air flow allowed to cease before the transducer offsets were logged. Immediately after these values were recorded, the appropriate data acquisition routine was initiated whilst the tunnel was brought up to speed and thence data gathered as per the software prompts. The tunnel was then shut down, offsets logged again and further tests were performed in the manner described above.

#### 3.5 Roughness Transition Strips

A number of the experiments were repeated with graded sand deposited at the aerofoil's leading edge. It was intended that this should trip the boundary layer in the leading-edge region. A direct comparison can be made between tests with and those without these roughness transition strips.

#### 3.6 Data Presentation

All data collected by the data acquisition routines were stored in unformatted form on magnetic tape. A library of programs (coded in FORTRAN 77) is available for the reduction, presentation and analysis of the data on a DEC MICROVAX 3400. By applying offsets, gains and calibrations, the data reduction programs convert the cycles of raw data into averaged or unaveraged non-dimensional pressure coefficients. As described by Leitch and Galbraith<sup>9</sup>, these data are stored on the University of Glasgow's aerofoil database. The airloads are determined by suitably integrating the pressure coefficient values.

#### **4 RESULTS AND DISCUSSION**

#### 4.1 Tunnel Performance

Assessment of the quality of the data can only be made with a clear insight of the tunnel effects. Unfortunately the tunnel performance was such that, for the time scales of the model motion, it was not possible to hold the dynamic pressure in the working section constant whilst altering the blockage due to the pitching of the aerofoil. During the static tests (i.e.  $\mathbf{k}=0.0$  and  $\mathbf{r}=0.0$ ), this variation was as illustrated in **Figure 5**, where it can be seen that there was approximately a 30% reduction in dynamic pressure as the angle of attack was increased from 0° to 30°. As illustrated in **Figures 6** and **7**, this reduction in dynamic pressure decreased as reduced frequency increased.

Figure 8 reveals that, during ramps, there was a drastic reduction and subsequent unsteadiness in the dynamic pressure during a test. The model was pitched to an incidence of  $40^{\circ}$  so that uniform ramp conditions existed at stall. Once the aerofoil had stalled, however,

all significant data had already been collected and the corresponding dynamic pressure reduction was only in the region of 10%. The subsequent data are of little relevance to the current work and is presented merely for completeness.

#### 4.2 Averaging of the Data

The main data in this report are the average of a number of cycles. Individual cycles are presented in Figures 9 and 10 where it may be seen that, whilst minor random differences do exist from cycle to cycle, the salient features are highlighted by the averaging process. In addition, the sweep at which any event occurred did not vary. Therefore the given data may be considered as typical of aerofoil performance during any given individual cycle. This is particularly relevant when considering the detailed flow phenomena of separation and reattachment.

#### 4.3 Test Data

The test data are grouped for each motion type with compact details of the specific tests listed in **Tables 2** to **5**.

#### ACKNOWLEDGEMENTS

The authors wish to acknowledge the encouragement and support of their colleagues both academic and technical.

The research was performed with funding from Vertical Axis Wind Turbines Limited (contract number **59220/0004D00**).

#### REFERENCES

<sup>1</sup>Galbraith, R.A.McD. and Leishman, J.G. A micro-computer based test facility for the investigation of dynamic stall. International Conference on the Use of Micros in Fluid Engineering, Paper E3, June 1983.

<sup>2</sup>Leishman, J.G. Contributions to the experimental investigation and analysis of aerofoil dynamic stall. Ph.D. Dissertation, University of Glasgow, United Kingdom, March 1984.

<sup>3</sup>Galbraith, R.A.McD. A data acquisition system for the investigation of dynamic stall. Proceedings of the 2<sup>nd</sup> International Conference on Computational Methods and Experimental Measurement. Computational Mechanics Centre Publication, Southampton, United Kingdom, 1984.

<sup>4</sup>Angell, R.K., Musgrove, P.J., Galbraith, R.A.McD. and Green, R.B. Summary of the collected data for tests on the NACA 0015, NACA 0018, NACA 0021, NACA 0025 and NACA 0030 aerofoils. Glasgow University Aero Report 9005, February 1990.

<sup>5</sup>Galbraith, R.A.McD, Barrowman, J. and Leishman, J.G. Description of the sample and hold circuits for the Glasgow University dynamic stall facility. Glasgow University Aero Report 8208, 1982.

<sup>6</sup>Murray-Smith, E. and Galbraith, R.A.McD. User manual for the Glasgow University unsteady aerodynamic facility software. Glasgow University Aero Report 8800, 1988.

<sup>7</sup>Sharpe, D.J. Vertical axis WECS design procedures I. Department of Aeronautical Engineering, Queen Mary College, University of London.

<sup>8</sup>Sheldahl, R.E. and Klimas, P.C. Aerodynamic characteristics of seven symmetrical airfoil sections through 180-degree angle of attack for use in aerodynamic analysis of vertical axis wind turbines. Sandia Laboratories Report SAND80-2144, 1981.

<sup>9</sup>Leitch, E. and Galbraith, R.A.McD. Guide to Glasgow University's aerofoil database. Glasgow University Aero Report 8700, 1987.

# TABLE 1 : AHAVAW AEROFOIL PROFILE AND COORDINATES



# Coordinates in %Chord

| Upper   | Surface  | Lower Surface |          |  |  |  |
|---------|----------|---------------|----------|--|--|--|
| Station | Ordinate | Station       | Ordinate |  |  |  |
| 0.000   | 0.000    | 0.000         | 0.000    |  |  |  |
| 0.081   | 0.643    | 0.081         | -0.643   |  |  |  |
| 0.324   | 1.270    | 0.324         | -1.270   |  |  |  |
| 0.729   | 1.868    | 0.729         | -1.868   |  |  |  |
| 1.295   | 2.458    | 1.295         | -2.458   |  |  |  |
| 2.021   | 3.074    | 2.021         | -3.074   |  |  |  |
| 2.906   | 3.713    | 2.906         | -3.713   |  |  |  |
| 3.948   | 4.366    | 3.948         | -4.366   |  |  |  |
| 5.146   | 5.027    | 5.146         | -5.027   |  |  |  |
| 6.498   | 5.685    | 6.498         | -5.685   |  |  |  |
| 8.002   | 6.333    | 8.002         | -6.333   |  |  |  |
| 9.655   | 6.962    | 9.655         | -6.962   |  |  |  |
| 11.454  | 7.565    | 11.454        | -7.565   |  |  |  |
| 13.398  | 8.132    | 13.398        | -8.132   |  |  |  |
| 15.481  | 8.656    | 15.481        | -8.656   |  |  |  |
| 17.702  | 9.128    | 17.702        | -9.128   |  |  |  |
| 20.056  | 9.539    | 20.056        | -9.539   |  |  |  |
| 22.539  | 9.882    | 22.539        | -9.882   |  |  |  |
| 25.149  | 10.153   | 25.149        | -10.153  |  |  |  |
| 27.880  | 10.348   | 27.880        | -10.348  |  |  |  |
| 30.728  | 10.464   | 30.728        | -10.464  |  |  |  |
| 33.688  | 10.500   | 33.688        | -10.500  |  |  |  |
| 36.756  | 10.454   | 36./56        | -10.454  |  |  |  |
| 39.926  | 10.330   | 39.926        | -10.330  |  |  |  |
| 43.194  | 10.132   | 43.194        | -10.132  |  |  |  |
| 46.553  | 9.863    | 40.553        | -9.803   |  |  |  |
| 50.000  | 9.528    | 50.000        | -9.320   |  |  |  |
| 53.528  | 9.129    | 53.520        | -9.129   |  |  |  |
| 57.131  | 8.009    | 57.151        | -0.009   |  |  |  |
| 60.803  | 0.130    | 64 530        | -7 577   |  |  |  |
| 04.539  | 6.052    | 68 333        | -6.952   |  |  |  |
| 08.333  | 6.932    | 72 179        | -6 277   |  |  |  |
| 12.178  | 0.211    | 76.068        | -5.554   |  |  |  |
| 70.008  | 1 792    | 70.008        | -4 783   |  |  |  |
| 92.050  | 3.067    | 83 050        | -3.967   |  |  |  |
| 87 0/6  | 3.104    | 87 946        | -3.104   |  |  |  |
| 01.940  | 2 103    | 91 953        | -2.193   |  |  |  |
| 05 072  | 1 232    | 95 973        | -1.232   |  |  |  |
| 100.000 | 0.220    | 100.000       | -0.220   |  |  |  |

### TABLE 2 : DETAILS OF STATIC TESTS

# TABLE 2.1 : SUMMARY OF STATIC TESTS (nominal)

| Reynolds Number | 0.8x10 <sup>6</sup> | 1.1x10 <sup>6</sup> | 1.5x10 <sup>6</sup> | 2.0x10 <sup>6</sup> |
|-----------------|---------------------|---------------------|---------------------|---------------------|
| Angle of Attack |                     | -2° to              | o 30°               |                     |

(all permutations)

### TABLE 2.2 : LIST OF STATIC TESTS (actual)

| Run     | Start | Sweep | Reynolds               |
|---------|-------|-------|------------------------|
| Number  | ര     | (°)   | No. x 10 <sup>-6</sup> |
| 00011   | -2    | 32    | 1.60                   |
| 00491   | -2    | 32    | 1.54                   |
| 00751   | -2    | 32    | 1.04                   |
| 00801   | -2    | 32    | 2.06                   |
| 01681   | -2    | 32    | 1.53                   |
| 03541   | -2    | 32    | 1.10                   |
| 03711   | -2    | 32    | 1.59                   |
| 03961   | -2    | 32    | 2.02                   |
| 04131   | -2    | 32    | 0.87                   |
| 04211   | -2    | 32    | 1.21                   |
| 04291   | -2    | 32    | 1.63                   |
| 04371   | -2    | 32    | 2.08                   |
| 04801   | -2    | 32    | 0.86                   |
| 04881   | -2    | 32    | 1.20                   |
| 05031   | -2    | 32    | 2.09                   |
| 05181   | -2    | 32    | 1.66                   |
| *805541 | -2    | 32    | 1.57                   |
| *805741 | -2    | 32    | 1.53                   |

(\*experiments with roughness transition strips)

# TABLE 3 : DETAILS OF SINUSOIDAL EXPERIMENTS

### TABLE 3.1 : SUMMARY OF SINUSOIDAL EXPERIMENTS AT FIXED REDUCED FREQUENCY (nominal)

| Mean Angle        | 0°                  |       |        |       |                     |       |                     |  |  |
|-------------------|---------------------|-------|--------|-------|---------------------|-------|---------------------|--|--|
| Amplitude         | 5.4°                | 10.0° | 12.2°  | 13.8° | 17.4°               | 22.6° | 32.8°               |  |  |
| Reduced Frequency | 0.05                |       |        |       |                     |       |                     |  |  |
| Reynolds Number   | 0.8x10 <sup>6</sup> |       | 1.1x10 | 6 ]   | 1.5x10 <sup>6</sup> |       | 2.0x10 <sup>6</sup> |  |  |

(all permutations)

## TABLE 3.2 : SUMMARY OF SINUSOIDAL EXPERIMENTS AT FIXED REYNOLDS NUMBER (nominal)

| Mean Angle        | 0°                  |       |       |       |       |       |       |  |  |
|-------------------|---------------------|-------|-------|-------|-------|-------|-------|--|--|
| Amplitude         | 5.4°                | 10.0° | 12.2° | 13.8° | 17.4° | 22.6° | 32.8° |  |  |
| Reduced Frequency | 0.02                | 0.    | .04   | 0.05  | 0.    | 06    | 0.075 |  |  |
| Reynolds Number   | 1.5x10 <sup>6</sup> |       |       |       |       |       |       |  |  |

(all permutations; tests at reduced frequency of 0.075 were repeated with roughness transition strips)

# TABLE 3.3 : LIST OF SINUSOIDAL EXPERIMENTS (actual)

| Run    | Mean | Amp'ude      | Reduced   | Reynolds               |
|--------|------|--------------|-----------|------------------------|
| Number | (°)  | $(^{\circ})$ | Frequency | No. x 10 <sup>-6</sup> |
| 14141  | 0    | 5.4          | 0.047     | 0.87                   |
| 14151  | 0    | 10.0         | 0.047     | 0.87                   |
| 14161  | 0    | 12.2         | 0.047     | 0.87                   |
| 14171  | 0    | 13.8         | 0.047     | 0.87                   |
| 14181  | 0    | 17.4         | 0.047     | 0.87                   |
| 14191  | 0    | 22.6         | 0.047     | 0.87                   |
| 14201  | 0    | 32.8         | 0.047     | 0.87                   |
| 14221  | 0    | 5.4          | 0.047     | 1.20                   |
| 14231  | 0    | 10.0         | 0.047     | 1.20                   |
| 14241  | 0    | 12.2         | 0.047     | 1.20                   |
| 14251  | 0    | 13.8         | 0.047     | 1.20                   |
| 14261  | 0    | 17.4         | 0.047     | 1.20                   |
| 14271  | 0    | 22.6         | 0.047     | 1.20                   |
| 14281  | 0    | 32.8         | 0.047     | 1.20                   |
| 14301  | 0    | 5.4          | 0.047     | 1.60                   |
| 14311  | 0    | 10.0         | 0.047     | 1.60                   |
| 14321  | 0    | 12.2         | 0.047     | 1.60                   |

|   | Run     | Mean | Amp'ude | Reduced   | Reynolds               |      |
|---|---------|------|---------|-----------|------------------------|------|
|   | Number  | (°)  | (°)     | Frequency | No. x 10 <sup>-6</sup> |      |
|   | 14331   | 0    | 13.8    | 0.047     | 1.60                   |      |
|   | 14341   | 0    | 17.4    | 0.047     | 1.59                   |      |
|   | 14351   | 0    | 22.6    | 0.047     | 1.59                   |      |
|   | 14361   | 0    | 32.8    | 0.047     | 1.59                   |      |
|   | 14381   | 0    | 5.4     | 0.049     | 2.07                   |      |
|   | 14391   | 0    | 10.0    | 0.049     | 2.06                   |      |
|   | 14401   | 0    | 12.2    | 0.049     | 2.05                   |      |
|   | 14411   | 0    | 13.8    | 0.049     | 2.05                   |      |
|   | 14421   | 0    | 17.4    | 0.049     | 2.04                   |      |
|   | 14431   | 0    | 22.0    | 0.049     | 2.04                   |      |
|   | 14442   | 0    | 52.8    | 0.049     | 1.98                   |      |
|   | 14451   | 0    | 10.0    | 0.019     | 1.50                   |      |
|   | 14401   | 0    | 12.2    | 0.019     | 1.50                   |      |
|   | 14481   | Ő    | 13.8    | 0.019     | 1.57                   |      |
|   | 14491   | 0    | 17.4    | 0.019     | 1.57                   |      |
|   | 14501   | 0    | 22.6    | 0.019     | 1.57                   |      |
|   | 14511   | ŏ    | 32.8    | 0.019     | 1.56                   |      |
|   | 14521   | Õ    | 5.4     | 0.038     | 1.56                   |      |
|   | 14531   | 0    | 10.0    | 0.038     | 1.56                   |      |
|   | 14541   | 0    | 12.2    | 0.038     | 1.56                   |      |
|   | 14551   | 0    | 13.8    | 0.038     | 1.56                   |      |
|   | 14561   | 0    | 17.4    | 0.038     | 1.56                   |      |
|   | 14571   | 0    | 22.6    | 0.038     | 1.55                   |      |
|   | 14581   | 0    | 32.8    | 0.038     | 1.55                   |      |
|   | 14591   | 0    | 5.4     | 0.047     | 1.56                   |      |
|   | 14601   | 0    | 10.0    | 0.047     | 1.56                   | 8.00 |
|   | 14611   | 0    | 12.2    | 0.047     | 1.55                   |      |
|   | 14621   | 0    | 13.8    | 0.047     | 1.55                   | =    |
|   | 14631   | 0    | 17.4    | 0.047     | 1.55                   |      |
| - | 14641   | 0    | 22.6    | 0.047     | 1.55                   |      |
|   | 14651   | 0    | 32.8    | 0.047     | 1.55                   |      |
|   | 14661   | 0    | 5.4     | 0.057     | 1.55                   |      |
|   | 140/1   | 0    | 10.0    | 0.057     | 1.55                   |      |
|   | 14001   | 0    | 12.2    | 0.057     | 1.55                   |      |
|   | 14091   | 0    | 15.0    | 0.057     | 1.55                   |      |
|   | 14701   | 0    | 22.6    | 0.057     | 1.55                   |      |
|   | 14721   | 0    | 32.8    | 0.057     | 1.55                   |      |
|   | 14731   | Ő    | 5.4     | 0.071     | 1.55                   |      |
|   | 14741   | Õ    | 10.0    | 0.071     | 1.55                   |      |
|   | 14751   | Ő    | 12.2    | 0.071     | 1.54                   |      |
|   | 14761   | 0    | 13.8    | 0.071     | 1.54                   |      |
|   | 14771   | 0    | 17.4    | 0.071     | 1.54                   |      |
|   | 14781   | 0    | 22.6    | 0.071     | 1.54                   |      |
|   | 14791   | 0    | 32.8    | 0.071     | 1.54                   |      |
|   | *815671 | 0    | 5.4     | 0.078     | 1.49                   |      |
|   | *815681 | 0    | 10.0    | 0.078     | 1.48                   |      |
|   | *815691 | 0    | 12.2    | 0.078     | 1.48                   |      |
|   | *815701 | 0    | 13.8    | 0.078     | 1.48                   |      |
| 1 | *815711 | 0    | 17.4    | 0.078     | 1.48                   |      |
|   | *815721 | 0    | 22.6    | 0.078     | 1.48                   |      |
|   | *815731 | 0    | 32.8    | 0.078     | 1.48                   |      |

### TABLE 3.3 : LIST OF SINUSOIDAL EXPERIMENTS (concluded)

\*experiments with roughness transition strips

#### **AHAVAW - VOLUME III**

# TABLE 4 : DETAILS OF SINGLE STREAMTUBE VAWT<br/>EXPERIMENTS

### TABLE 4.1 : SUMMARY OF VAWT EXPERIMENTS AT FIXED REDUCED FREQUENCY (nominal)

| Mean Angle        | 0°                  |      |        |                        |      |      |                   |  |  |  |
|-------------------|---------------------|------|--------|------------------------|------|------|-------------------|--|--|--|
| Tip Speed Ratio   | 1.75                | 2.33 | 2.80   | 3.25                   | 3.50 | 4.00 | 6.00              |  |  |  |
| Reduced Frequency | 0.05                |      |        |                        |      |      |                   |  |  |  |
| Reynolds Number   | 0.8x10 <sup>6</sup> |      | 1.1x10 | 1.1x10 <sup>6</sup> 1. |      | 2.0  | 0x10 <sup>6</sup> |  |  |  |

(all permutations)

### TABLE 4.2 : SUMMARY OF VAWT EXPERIMENTS AT FIXED REYNOLDS NUMBER (nominal)

| Mean Angle        | 0°            |    |      |                     |      |      |       |  |  |
|-------------------|---------------|----|------|---------------------|------|------|-------|--|--|
| Tip Speed Ratio   | 1.75 2.33 2.8 |    | 2.80 | 3.25                | 3.50 | 4.00 | 6.00  |  |  |
| Reduced Frequency | 0.02          | 0. | 04   | 0.05                | 0.   | 06   | 0.075 |  |  |
| Reynolds Number   |               |    |      | 1.5x10 <sup>6</sup> | 5    |      |       |  |  |

(all permutations)

# TABLE 4.3 : LIST OF VAWT EXPERIMENTS (actual)

| Run    | Mean |      | Reduced   | Reynolds               |
|--------|------|------|-----------|------------------------|
| Number | (°)  | TSR  | Frequency | No. x 10 <sup>-6</sup> |
| 54811  | 0    | 6.00 | 0.047     | 0.86                   |
| 54821  | 0    | 4.00 | 0.047     | 0.86                   |
| 54831  | 0    | 3.50 | 0.047     | 0.86                   |
| 54841  | 0    | 3.25 | 0.047     | 0.86                   |
| 54851  | 0    | 2.85 | 0.047     | 0.86                   |
| 54861  | 0    | 2.33 | 0.047     | 0.86                   |
| 54871  | 0    | 1.75 | 0.047     | 0.86                   |
| 54891  | 0    | 6.00 | 0.046     | 1.20                   |
| 54901  | 0    | 4.00 | 0.046     | 1.20                   |
| 54911  | 0    | 3.50 | 0.046     | 1.19                   |
| 54921  | 0    | 3.25 | 0.046     | 1.19                   |
| 54931  | 0    | 2.85 | 0.046     | 1.19                   |
| 54941  | 0    | 2.33 | 0.046     | 1.19                   |
| 54951  | 0    | 1.75 | 0.046     | 1.19                   |

| Run    | Mean |      | Reduced   | Reynolds               |
|--------|------|------|-----------|------------------------|
| Number | (°)  | TSR  | Frequency | No. x 10 <sup>-6</sup> |
| 54961  | 0    | 6.00 | 0.046     | 1.60                   |
| 54971  | 0    | 4.00 | 0.046     | 1.60                   |
| 54981  | 0    | 3.50 | 0.046     | 1.60                   |
| 54991  | 0    | 3.25 | 0.046     | 1.59                   |
| 55001  | 0    | 2.85 | 0.046     | 1.59                   |
| 55011  | 0    | 2.33 | 0.046     | 1.59                   |
| 55021  | 0    | 1.75 | 0.046     | 1.58                   |
| 55041  | 0    | 6.00 | 0.046     | 2.07                   |
| 55051  | 0    | 4.00 | 0.046     | 2.07                   |
| 55061  | 0    | 3.50 | 0.046     | 2.06                   |
| 550/1  | 0    | 3.25 | 0.045     | 2.05                   |
| 55081  | 0    | 2.85 | 0.045     | 2.05                   |
| 55091  | 0    | 2.33 | 0.045     | 2.04                   |
| 55102  | 0    | 1.75 | 0.049     | 1.97                   |
| 55111  | 0    | 6.00 | 0.018     | 1.57                   |
| 55121  | 0    | 4.00 | 0.018     | 1.57                   |
| 55131  | 0    | 3.50 | 0.018     | 1.56                   |
| 55141  | 0    | 3.25 | 0.018     | 1.56                   |
| 55151  | 0    | 2.85 | 0.018     | 1.56                   |
| 55101  | 0    | 2.33 | 0.018     | 1.56                   |
| 551/1  | 0    | 1./5 | 0.018     | 1.56                   |
| 55191  | 0    | 6.00 | 0.037     | 1.64                   |
| 55201  | 0    | 4.00 | 0.037     | 1.64                   |
| 55211  | 0    | 3.50 | 0.037     | 1.64                   |
| 55221  | 0    | 3.25 | 0.037     | 1.64                   |
| 55241  | 0    | 2.85 | 0.037     | 1.64                   |
| 55251  | 0    | 2.33 | 0.037     | 1.03                   |
| 55251  | 0    | 1.75 | 0.037     | 1.03                   |
| 55271  | 0    | 4.00 | 0.040     | 1.04                   |
| 55291  | 0    | 4.00 | 0.040     | 1.04                   |
| 55201  | 0    | 2.25 | 0.040     | 1.05                   |
| 55301  | 0    | 2.25 | 0.040     | 1.05                   |
| 55311  | 0    | 2.05 | 0.040     | 1.05                   |
| 55321  | 0    | 1 75 | 0.040     | 1.62                   |
| 55331  | 0    | 6.00 | 0.040     | 1.61                   |
| 55341  | 0    | 4.00 | 0.055     | 1.61                   |
| 55351  | 0    | 3 50 | 0.055     | 1.61                   |
| 55361  | 0    | 3 25 | 0.055     | 1.60                   |
| 55371  | õ l  | 2.85 | 0.055     | 1.60                   |
| 55381  | 0 I  | 2 33 | 0.055     | 1.60                   |
| 55391  | 0    | 1.75 | 0.055     | 1.60                   |
| 55401  | 0    | 6.00 | 0.069     | 1.61                   |
| 55411  | õ l  | 4 00 | 0.069     | 1.61                   |
| 55421  | 0    | 3 50 | 0.069     | 1.60                   |
| 55431  | 0 I  | 3 25 | 0.069     | 1.60                   |
| 55441  | 0    | 2.85 | 0.069     | 1.60                   |
| 55451  | 0 I  | 2 33 | 0.069     | 1.60                   |
| 55461  | ŏ    | 1.75 | 0.069     | 1.60                   |

### TABLE 4.3 : LIST OF VAWT EXPERIMENTS (concluded)

#### AHAVAW - VOLUME III

# TABLE 5 : LIST OF NON-STANDARD VAWT EXPERIMENTS

| Run                | Mean | Amp'ude | Reduced   | Reynolds               |
|--------------------|------|---------|-----------|------------------------|
| Number             | (°)  | Ô       | Frequency | No. x 10 <sup>-6</sup> |
| <sup>a</sup> 55482 | 0    | 12      | 0.045     | 2.00                   |
| <sup>b</sup> 55492 | 0    | 12      | 0.045     | 1.99                   |
| °55502             | 0    | 20      | 0.045     | 1.98                   |
| d55512             | 0    | 20      | 0.045     | 1.97                   |
| e55521             | 0    | 30      | 0.046     | 1.96                   |
| f55531             | 0    | 30      | 0.046     | 1.94                   |

<sup>a</sup>angle of attack trace from simulation of the VAWT 260 at a wind speed of 7.89 ms<sup>-1</sup> (structural dynamic effects not included)

<sup>b</sup>angle of attack trace from simulation of the VAWT 260 at a wind speed of 7.89 ms<sup>-1</sup> (structural dynamic effects included)

<sup>c</sup>angle of attack trace from simulation of the VAWT 260 at a wind speed of 10.57 ms<sup>-1</sup> (structural dynamic effects not included)

<sup>d</sup>angle of attack trace from simulation of the VAWT 260 at a wind speed of 10.57 ms<sup>-1</sup> (structural dynamic effects included)

eangle of attack trace from VAWT 260

<sup>f</sup>equivalent angle of attack trace for wind tunnel to reproduce airloads from VAWT 260







7ft X 5ft Jin WIND TUNNEL



FIGURE 3: PRESSURE TRASDUCER LOCATIONS FOR THE AHAVAW.



SYSTEM SYSTEMATIC ABRANGEMENT OF DATA ACOUSITION AND CONTROL FIGURE 4:

•











FIGURE 9: EFFECT OF AVERAGING ON THE NORMAL FORCE AND PITCHING MOMENT FOR OSCILLATORY TESTS.



FIGURE 10: TYPICAL UNAVERAGED DATA FOR RAMP TESTS.

G.U. Aero Report 9214

# **UNIVERSITY OF GLASGOW**

# **DEPARTMENT OF AEROSPACE ENGINEERING**

PRESSURE DATA FROM STATIC EXPERIMENTS



DYNAMIC PRESSURE =  $1121.96 \text{ Nm}^2$ REYNOLDS NUMBER = 1599672. RUN REFERENCE NUMBER: NUMBER OF CYCLES = 1

og 81.804 mN d'

x/c

-21



49

-0.3L

16.0-

x/c

2.0 000

C C C





DYNAMIC PRESSURE =  $1801.81 \text{ Nm}^{-2}$ RUN REFERENCE NUMBER: 801 REYNOLDS NUMBER = 2057753. NUMBER OF CYCLES = 1

> ar an env. G 0 PO

0.1 0.6 0.8 X/C



168

- Do

0 N O D T O N

d'



×

354

0 P P

d'

0 1 0 1



4-04 N

0 N 0 0 4



15.83 11.2 6.61 -2.08 0.2 0.4 0.6 0.8 1 α Δ L - P0

alpha

x/c

396



ar an t an t

PO



₽ ₽ ₽

alpha

x/c

421

-0.3

-0.J



P P P 000



0.2 

- Po


Ş



DYNAMIC PRESSURE =  $655.01 \text{ Nm}^{-2}$ RUN REFERENCE NUMBER: 4881 REYNOLDS NUMBER = 1200616. NUMBER OF CYCLES = 1 488

0.4 0.6 0.8 x/c

ar.

do-

0 N 0 0 F



 -2



- 10 m

-Cp 3

à r



10



Cps at LE, TE, 30%

8

40,

DYNAMIC CHARACTERISTICS FOR THE AHAVAW - VAWT Model

DYNAMIC PRESSURE = 1059.48 Nm<sup>-2</sup> RUN REFERENCE NUMBER: 805541 REYNOLDS NUMBER = 1567557. NUMBER OF CYCLES = 1

DATE OF TEST: 4/1/92 MACH NUMBER = 0.122

ANGLE





-2.07 0.6 0.8 200-

00 - 00 - 00 - 00

alpha

X/C

554

-0.3L -0.2

ť

ct -0.2 -0.31



-CP

81.62

574

•

G.U. Aero Report 9214

## **UNIVERSITY OF GLASGOW**

## **DEPARTMENT OF AEROSPACE ENGINEERING**

PRESSURE DATA FROM

## **OSCILLATORY EXPERIMENTS**



-CP 3.

8 × 6 10 4

000



-2

-Cp 2

0000



ar an env. av

- P0

415

;



, A A

oq 2



0000 F 00

000

do-











11.24 5.08 -2.08 alpha

مرح مرم جر هر ح م ا

0.4 0.6 0.8 x/c

000-



٦

-0.JE. 0-

-0.3

-0.2

ئ

-0.1

-20 -10



0 NON TON

000

e.



OMEGA x T (rads.



18



12

484

16.0--0.2

-0.3L

alpha

0.4 0.6 X/C

A POLY A POLY

ť





Cps at LE, TE, 30%





ANGLE OF ATTACK



16.72 15.25 6.65

\* 10 0 A 00

-2.74 alpha

> 0.4 0.6 0.8 x/c

00

ę

2



٩

485



-0.3 -0.2 -0-

ť



0.2 0.4 0.6 0.8 0 N O S T O N 00 N do-

7.05 -1.73

alpha

x/c



| RUN REFERENCE NUMBER: 54861                | DATE OF TEST: 12/11/91                        |
|--|---|
| REYNOLDS NUMBER = 861679.                  | MACH NUMBER = $0.070$                         |
| DYNAMIC PRESSURE = 334.25 Nm <sup>-2</sup> | AIR TEMPERATURE = 19.5                        |
| NUMBER OF CYCLES = 10                      | SAMPLING FREQUENCY =                          |
| MOTION TYPE: VAWT FUNCTION                 | REDUCED FREQUENCY = 0                         |
| MEAN ANGLE = 0.00°                         | <b>AMPLITUDE = <math>22.60^{\circ}</math></b> |
| OSCILLATION FREQUENCY = 0.650 Hz.          |   |
| AVERAGED DATA OF 10 CYCLES                 |   |





0.5

1001-

30

-1.5

10

OMEGA x T (rads.)

0

2.0<sup>r</sup>

....

ۍ د





21.43

13.3 0.71

-12.02 7-20.4

-19. J

21.80 18.60







486

...

Cps at LE, TE, 30%

8

10

-0.3

486

0.2 0.4 0.6 0.8 x/c

7.8 -3.18, alpha



BURNEWN-04 ę.



Cps at IE, TE, 308 MEGA X



ANGLE OF ATTACK -2.01



487

-0.3

-0.3

-3.82, alpha

> 0.4 0.6 0.8 x/c

Po -21

0000

BN

0-



Po Po



.

489

16.0-

-0.2

ىلى

Ct -0.2

-0.1

-1.32

alpha

0.1 0.6 0.8 x/c

-0.3

-0.1



- 00 - B

-2 PO

-Cp 2



-0.3

00

-21

-Cp 3

8000



Pog-

0 0 4 0 0 



0

491

-0.3L

-0.2

പ്

ct -0.2

-0.3

4 6 6 6 7

7 -2.99

.2 0.4 0.6 0.8 x/c





Cps at LE, TE, 30%

492







13.7 12.1.

0000

-3.26, alpha

> 0.4 0.6 0.8 x/c

2 p

dy-



-2.0

ANGLE OF ATTACK

0.1

...

4.0-0

10

-0.3L

-0.3

-0.2

ť



8 × 6 0 × 0 0 d' alpha

x/c

-2 PO

426

-0.3

-0.31







-0.JE

Cps at LE, TE, 30%

8

10

493

OMEGA x T (rads.)

5



]6

-2.0

-

ANGLE OF ATTACK



1.0 2.0 3.0 4.0 5.0 8.0 10:0 -0.J 0.1

0.2 -0.1

ct -0.2

4 0.6 0.8 X/C - CD - CD - CD - CD 00

-3.88 alpha

17.43 14.68 5.53





.

| RUN REFERENCE NUMBER: 54941                 | DATE   |
|---|--------|
| REYNOLDS NUMBER = 1191777.                  | MACH N |
| DYNAMIC PRESSURE = $656.17 \text{ Nm}^{-2}$ | AIR TE |
| NUMBER OF CYCLES = 10                       | SAMPLI |
| MOTION TYPE: VAWT FUNCTION                  | REDUCE |
| MEAN ANGLE = 0.00°                          | AMPLIT |
| OSCILLATION FREQUENCY = 0.894 Hz.           |        |
| AVERAGED DATA OF 10 CYCLES                  |        |

 oF TEST: 13/11/91
 1
 20 

 I NUMBER = 0.098
 34
 0

 TEMPERATURE = 22.7°C
 0.0
 0

 JING FREQUENCY = 114.43 Hz.
 -20
 0

 USED FREQUENCY = 0.046
 -40
 -40

 JIUDE = 22.60°
 0.046
 -40



OMEGA x T (rads.)

12 0.

8- 00-





ANGLE OF ATTACK



494





494

Cps at LE, TE, 30%

8

40-



•

-0q

do-

-2

87.004 BN



8 × 9 0 4

-Cp 3-

PQ N



-Cp 3

8 65 000-


x/c

E.0-





497

ct -0.2

497

-0.3L

16.0-

-1.45 alpha

> 0.2 0.4 0.6 0.8 X/C

000

-Cp 3

2018

-0.2

ىلى



<del>۵۲.۵۵۴.۴۵۷-</del> 



498

B C C C

DAN



00 N



۳. میں <u>د سر م</u> م ا



۲ ۲ ۳ ۳ 00 N



500

-0.31

-0.3

аголено-б 1 Da ?



435

8

004 NN oq

-cb

-2



8 9 9 9 9 9 9 9 9 9 9 9 9

alpha

X/C

501

-0.3

-0.3





0-0-0PO N



P P P





NUL BUIL

-09 "

-cb



Poor N

-2004-00-



2014 20 1 B

-cb

00 N





00 N

ę

Bront mn



P P P



0 N M P N N A M

000

d'



508

-0.3

-0.3

alpha

0.4 0.6 0.8 x/c

og v

-cp 2



00 N

do-

Brunt mo



P P

N PO





-0.J

ę



445

Par-

Brone Br

-cb



-cp 2.

8000

og -2



446

-0.31

-0.3

x/c

-2-

pa

-Cp 3



OMEGA X T (rads.)



-0-

512

-0.3 -0.2

പ

ct -0.2

-1.23 alpha

> 0.2 0.4 0.6 0.8 x/c

d-o-

0000

-0.1

-0.3



Per n

-cp 2



OMEGA x T (rads.)



ANGLE OF ATTACK

10



ANGLE OF ATTACK



513

-0.3



20.0

11.57 5.81 -1.39

000

alpha

0.2 0.4 0.6 0.8 x/c

00 -21

dy-

ct -0.2

-0.1

-0.3

513

Cps at LE, TE, 30%



448

-0.3

-0.3

x/c

og 2

d'



OMEGA x T (rads.)







12.95 6.38

-1.57 alpha

> 0.4 0.6 0.8 X/C

Poq 2

-CP 2.

-

00 10

ANGLE OF ATTACK

-0.3

-20 -10

-0.2

514







449

10

ct -0.2 -0.31

-0.93/

alpha

0.2 0.4 0.6 0.8 x/c

000

00004

mn do-

-0.3

-0.2

ť








ANGLE OF ATTACK

0.1

-0.1

-24-CLa



16.57 15.73 7.52 -1.82

0000

alpha

0.4 0.6 0.8 x/c

200

<u>ч</u> Ч









515

-0.31

515

Cps at LE, TE, 30%



450

00 N

-Cp

0 N 0 0 4









ANGLE OF ATTACK







0.1

-0.1



516



451

-0.3

ť

ct -0.2

-1.51

alpha

0.2 0.4 0.6 0.8 x/c

- PO

60TON

dy-

10

-0.31















ct -0.2

-0.1

10.59

0000

-2.66 alpha

> 0.2 0.4 0.6 0.8 X/C

000

do-

16.0-



0000

-2 q

dy-



16.0-

x/c

9.

0000

nn

d'



NG

-2

00TON

do-







ANGLE OF ATTACK





-20 -10

1.0 2.6 3.0 4.0 5.0

ť

-0.2

ct -0.2 (0°.0) -0.1 0.1



-0.1

520

-0.3

-0.3

X/C

-2



]6

-CP -CP

85

-2











-0.1



ೆ

16.0-

521

10.7 3.96

-4.00 alpha

> 0.4 0.6 0.8 x/c



NUM + UIG 1 B

dy-







ANGLE OF ATTACK







522

522

-0.31

-0.3

-0.

12.05

8 6 6 5

-4.51 alpha

> 0.4 0.6 0.8 x/c

97

d'

























17.50 14.51 5.16 -5.36 alpha

85 60 0.4 0.6 0.8 x/c

N PO

-Cp 3.

-A0:0 ct -0.2 0.1 -0.1

-0.3

10.0-

523















523

P



00 -2

n'n dy-













524

-0.3

E. 0-

-6.34

alpha

0.4 0.6 0.8 x/c

ġ

-cp

0 1 9 10





ar-

-Cp 2.



-2

PG

d'o'



-Cp





527

-0.3

-21

24

-cp





-0.3

x/c

2

-Cp 3

0000



-CP 2

8 in in 00 N





529

NO NO

-cp 2



463

0000

FON P do-



]

-0.3

530

-0.3

alpha

X/C

- PO

-CP 2.



nn

do-





P. 0 0 4 0 0 .

- PO



q -2





CP + 0.0 - 0.0 - CP



P P P P

- PO


534

-0.31

-0.3

100

-Cp 3.



nn

-cb

18 6 5 -2

PG



5

535

-0.3

-0.3

alpha

0.1 0.6 0.8 x/c

dy-



.

م م م م م م م م م م م

0 P 0



Par-

-CP 2



-14.07

-7.48

14.26

16.77

15.51 10.45

0000

-4.37 alpha

> 0.4 0.6 0.8 x/c

PG

d'o-

10

470

10



OSCILLATION FREQUENCY = 1.462 Hz. DYNAMIC PRESSURE = 1139.07 Nm<sup>-2</sup> RUN REFERENCE NUMBER: 14701 AVERAGED DATA OF 10 CYCLES REYNOLDS NUMBER = 1549348. MOTION TYPE: SINUSOIDAL NUMBER OF CYCLES = 10 0.00 MEAN ANGLE =



537

G N

do-



-5.74

alpha

x/c

-2

471

-0.31

-0.2

ť

ct -0.2

-0.3















004 m N -cb





200

-Cp 2

9 5



nn

-Cp

2-P



-21

PG

g

0000



Co Co Co Co Co

à -2



542

-0.3

-0.3

x/c

og ?

d'

87.604 mN





]? 30 476

-0.31

-0.31

x/c

2

-Cp 3.







543

-0.3

-0.3

alpha

X/C

-2

Pq

0000

30

ð





-0.3

-0.3

x/c

N 10 2 00 1 00

do-

477



-0.3 -0.2 -0--20 -10 ť



-3.40 alpha

6 5 8

ę

x/c



30 40



Par

0000

-Cb -



P OMEGA X T (rads.)







E. 0-

-4.19 alpha

8000 T 00-

d' O

X/C

N PO





0 N U D T O O A O

oq -21

do-



- PO





P. C. C. C. C. C. De la



568

568

JE. 0-

10.0-

alpha

x/c

Po N

-00 2

0000-



x/c



10.0-

×/c

2 PO

۵۲-99-4-60 6



-0.3L

571

104

30

ct -0.2 -0.31

-5.71 alpha

0.4 0.6 0.8 1 x/c

og 2



18

572

-0.3L

-0.2

ں ٹ

ct \_0.2 -0.3

-7.44

alpha

0.6 0.8 1.

N PO-

X/C



20 30 40

0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0

og N



20 30

ANGLE OF ATTACK

ct -0.2 -0.3

-0.3 -0.2

പ്

alpha

0.4 0.6 0.8 x/c

2.0

Po Po



2.0 -00-1

NU440010

d'



550

-0.3

-0.3

å



00 -21

<sup>d</sup>

800540N


| RUN REFERENCE NUMBER: 55521                   |     |  |
|---|-----|--|
| REYNOLDS NUMBER = 1959005.                    |     |  |
| DYNAMIC PRESSURE = $1614.41$ Nm <sup>-2</sup> | N   |  |
| NUMBER OF CYCLES = 10                         |     |  |
| MOTION TYPE: VAWT FUNCTION                    |     |  |
| MEAN ANGLE = 0.00°                            |     |  |
| OSCILLATION FREQUENCY = 1.360                 | Hz. |  |

174.09 Hz. REDUCED FREQUENCY = 0.046 AIR TEMPERATURE = 18.5°C DATE OF TEST: 29/1/92 SAMPLING FREQUENCY = MACH NUMBER = 0.149AMPLITUDE =  $30.00^{\circ}$ 







0.0

-1.0

-1.5

-2.01

Cn 2.0

AVERAGED DATA OF 10 CYCLES

1.0 0.5





-2.0

-1.5







552

Cps at LE, TR 30%

8 1

(0



553

No. 1

å

87.004 mN

553

