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UNIVERSITY OF GLA

Engineering PERIODICALS

DEPARTMENT OF AEROSPACE ENGINEERING

COLLECTED DATA FOR TESTS ON A NACA 0015 AEROFOIL WITH CHORD OF LENGTH 0.275m.

by

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March 1992

COLLECTED DATA FOR TESTS ON A NACA 0015 AEROFOIL WITH CHORD OF LENGTH 0.275m.

Herein is presented the collected data for tests in which a NACA 0015 aerofoil with chord of length 0.275m, was subjected to a variety of displacements in pitch about the quarter-chord location at low Reynolds numbers.

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SHORTENED NACA 0015

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NOMENCLATURE

c	chord
Cm	pitching-moment coefficient
Cn	normal force coefficient
Cp	pressure coefficient
Ct	"thrust" force coefficient
D.P.	dynamic pressure ($\rho V^2/2$)
k	reduced frequency ($\omega c/2V$)
r	reduced pitch-rate (c/2V)do/dt
Re	Reynolds number
TSR	tip speed ratio
V	velocity
x/c	chordwise dimension
α	angle of attack
ω	rotational velocity

1 INTRODUCTION

The phenomenon of dynamic stall, the onset of which is largely controlled by the behaviour of the viscous boundary layer on the aerofoil surface, plays an important role in the successful design of the helicopter rotor. During high speed forward flight conditions, the blades on the retreating side of the rotor disc encounter a reduced dynamic pressure, and hence rotor trim requirements dictate a high aerofoil lift coefficient. These high lift coefficients are generated through large angles of incidence, often exceeding the maximum static stall value and so take advantage of the dynamic effects on the stalling process. Aerofoil dynamic stall is imprecisely understood and is currently the subject of extensive experimental and theoretical investigation by, amongst others, **Beddoes**¹. As has been shown by Harris and Pruyn², attempts to predict rotor performance without a mathematical model of retreating-blade stall have met with little success. Furthermore, the modelling is complicated by the highly threedimensional flowfield of the rotor. It is clear, however, that, in order to formulate modelling techniques for use in rotor airload calculations, a basic understanding of the unsteady stall process must be established.

An experimental investigation of retreatingblade stall, together with a boundary-layer analysis on a model rotor, by **McCroskey et al^{3,4}** pointed to the modelling of blade dynamic stall by an oscillating aerofoil in the nominally two-dimensional flow environment of a wind tunnel. Many such experiments of aerofoils oscillating through stall have since been performed, and data have been gathered for both the analysis of the fluid mechanics of the dynamic stall phenomenon itself and for use in mathematical model development.

As part of this investigation, in recent years, in the dynamic stall facility at the University of Glasgow^{5,6,7}, two-dimensional data have been acquired from experiments on a number of aerofoils under a variety of motion types. These aerofoils can be divided into two groups: the first is a family of cambered aerofoils generated from the NACA 23012 section and intended for the examination on helicopter blades of the transition from trailing-edge to leading-edge stall and the mechanism of reattachment; the second is a series of symmetrical sections for use on large-scale vertical-axis wind turbines. Of this second group of aerofoils, Angell et al^{8,9,10} tested the NACA 0015. That model, like all others tested before now, possessed a chord of length 0.55m. However, as part of the University's research into the effects on the data of model aspect ratio and tunnel blockage, it was decided to repeat a number of these experiments on a model of chord length 0.275m. This report presents the collected data from tests performed on this shortened NACA 0015 model in steady and unsteady conditions. The coordinates for the aerofoil section are listed in Table 1, and a brief descripton of the experimental apparatus and techniques is presented below.

2 DESCRIPTION OF TEST FACILITY

2.1 Aerofoil and Wind Tunnel

The general arrangement of the aerofoil in the wind tunnel was as shown in **Figure 1**. The aerofoil, of chord length 0.55m and span 1.61m, was constructed of fibre glass mounted on an aluminium spar and filled with an epoxy resin foam. The hand-finished surface was very smooth, and the profile accurate to better than 0.1mm. The instrumented model was fitted vertically into the University of Glasgow's "Handley Page" wind tunnel.

The "Handley Page" low-speed wind tunnel is an atmospheric-pressure closed-return type with a 1.61x2.13 octagonal working section (**Figure 2**) in which a wind velocity of $61ms^{-1}$ can be attained. The model was pivoted about

its quarter-chord axis on two tubular steel shafts connected to the main support via two selfaligning bearings. A single thrust bearing on the top support beam took all the weight. The dynamic and aerodynamic loadings from the aerofoil were reacted to the tunnel framework by two transversely mounted beams.

2.2 Pitch Drive Mechanism

2.2.1 Actuator

Angular movement of the model was obtained using a linear hydraulic actuator and crank mechanism. The actuator was mounted horizontally below the tunnel working section on the supporting structure, with the crank rigidly connected to the tubular part of the spar by a welded sleeve and keyway. The acuator was a UNIDYNE 907/1 type with a normal dyname thrust of 6.1KN operated from a supply pressure of 7.0MNm⁻². A MOOG 76 series 450 servo valve was used via a UNIDYNE servo controller unit to control the movement of the actuator. A suitable feedback signal for the controller was provided by a precision linear angular displacement transducer geared to the main spar of the model.

2.2.2 Command Signal

The model's angle of attack was incremented by the actuator controller. The input signal during the static tests was provided under software control by the data acquisition unit's own digital-to-analogue converter. This was possible because, during the sampling, the angle of attack was fixed and sufficient time was available between sampling to set the model at the required angle of attack. The two activities were separate and were performed sequentially.

Such was not the case during the unsteady tests, however, where sampling and control of the model's motion were required simultaneously. Therefore, during these tests, the input signal was provided by a separate function generator, comprised of an AMSTRAD 1512 microcomputer equipped with an ANALOG DEVICES RTI815 multi-function input/output board. The required output function was digitised into equal time steps in 2's complement code and the frequency of the function was controlled using the internal interrupts of the AMSTRAD microcomputer. The code was written in TURBO PASCAL.

2.3 Instrumentation and Data Logging

2.3.1 Pressure Transducers

To provide the chordwise pressure distribution at mid-span, thirty KULITE XCS-093-5 PSI G ultra-miniature pressure transducers were installed just below the surface of the centre section of the model. The transducers were of vented gauge type with one side of the pressure sensitive diaphragm open to the ambient pressure outside the wind-tunnel (via tubes in the model). Each transducer was fitted with a temperature compensation module, which minimised the change in zero-offset and sensitivity with temperature. The locations of the pressure transducers in the model are illustrated in **Figure 3**.

The low voltage outputs from the thirty presure transducers were suitably amplified and conditioned by a bank of differential amplifiers. The conditioned signals were passed to a "sample and hold" unit^{5,11} to overcome the time-skew problem arising from the sequential conversion of the anlogue signals into digital form.

2.3.2 Dynamic Pressure

The dynamic pressure in the wind tunnel working section was determined by measuring the difference between the static pressure in the working section, 1.2m upstream of the leading edge, and the static pressure in the settling chamber. The pressure tappings were connected to a FURNESS FC012 micromanometer, which provided an analogue signal suitable for the data acquisition unit's analogue-to-digital converter. This dynamic pressure was recorded as the sample-and-hold unit was triggered to sample the output from the pressure transducers.

2.3.3 Incidence

The instantaneous angle of attack of the aerofoil was determined by an angular displacement transducer geared to the model's main spar. The signal voltage from the transducer was fed into an amplifier/splitter to produce three signals for the following purposes:

- i) connection of the multiplexer for recording the aerofoil's angle of attack;
- ii) connection of the Schmitt trigger for initiation of data sampling when a preset incidence (voltage) was attained;
- iii) a feedback signal to the hydraulic actuator controller.

2.3.4 Acquisition Unit

The actual data acquisition unit was a DEC MINC-11 microcomputer, configured with an LSI-11/32 16-bit microprocessor and laboratory modules which included:

- i) an analogue-to-digital converter module, with a 16-channel multiplexer incorported. The converter was a 12-bit successive approximation type with a conversion time of of $30\mu s$, but the multiplexer's settling time and the need to transfer the data from the analogue-todigital converter into system memory increased the conversion time to $44\mu s$;
- ii) a multiplexer module, of 16 single-ended channels, which increased the number of channels that could be sampled to 32;
- iii) a real-time clock module, with two Schmitt triggers. This was used as a time-base generator to accurately set the sampling frequency. For ramp experiments, the sampling frequency was determined at run time from the pitch rate and the requirement that 128 sample sweeps should be obtained when the incidence was increasing and the same number when the aerofoil was sitting at its final incidence. However this specification was qualified by the fact that data were required to be recorded at the final incidence for no longer than 4 seconds and that the maximum sampling frequency which could be attained was 550Hz. One of the Shmitt triggers was used to initiate data sampling, by setting its reference voltage to a value corresponding to the angular

displacement transducer's output for the required starting angle of attack. For oscillatory tests, the sampling frequency was determined from the frequency of oscillation and the requirement that 128 sample sweeps should be obtained during each cycle;

iv) a digital-to-analogue converter module which housed four independent 12-bit digital to analogue converters. This was used to provide the command signal for the hydraulic actuator during static tests.

The path of data flow and system layout is shown diagrammatically in **Figure 4**. The main control programs for the tests were written in FORTRAN IV, as described by **Murray**-**Smith and Galbraith¹²**. The programs prompt the user for specific run information before calling a specialised subroutine written in MACRO-11 assembly language to receive and store the digitised data. The timing and control of the analogue-to-digital converter and associated circuitry was performed by the processor's hardware, but channel selection and data management were achieved under software control.

Part of the research on this model has involved calculating the velocity of the dynamic stall vortex over the aerofoil's upper surface. It was perceived that, for this purpose, it was necessary to record data at a higher frequency than the 550Hz which could be recorded by the system described above. Therefore, for a number of experiments, the data were converted by a THORN-EMI BE256 unit and logged via an IBM PSI180 microcomputer. These tests are indicated in the presentation plots by the fact that the sampling frequency was greater than 550Hz.

3 TEST SERIES AND PROCEDURE

3.1 Static Experiment

A number of experiments were performed under steady conditions. Once the wind velocity had reached the required value, the aerofoil was rotated about its quarter-chord axis until it was positioned at the incidence at which the first set of data were to be recorded.

Usually, this was approximately -2° . The model's angle of attack was then increased in steps of approximately 0.5° . After each increment in incidence, the flow was allowed to stabilise for a few seconds before each transducer's output was sampled 100 times and the mean value for each was stored. After 64 sweeps of data had been recorded, the model was returned to its starting position. Data sampling was maintained at the same rate on the return arc in order to record any delay in the reattachment of flow.

3.2 Unsteady Static Experiment

To record any fluctuations in the post-stall characteristics, the model's angle of attack was set at the required value and then a continuous sample of 256 data sweeps were recorded at a preset frequency (100Hz and 500Hz).

3.3 Ramp Experiment

During a ramp test the aerofoil was rotated about its quarter-chord axis over a preset arc at a constant pitch-rate. Five cycles of 256 data sweeps were recorded during each experiment. Between each ramp, the model sat at the finishing angle for five seconds, moved smoothly back to the starting angle in five seconds and sat at this position for another five seconds. Experiments were performed both when the pitch-rate was positive ("ramp up") and when it was negative ("ramp down").

3.4 Sinusoidal Experiment

For this experiment, the model was rotated about its quarter-chord axis so that its angle of attack varied sinusoidally with time. The amplitude and frequency were controlled by the AMSTRAD function generator. During each oscillatory cycle 128 data sweeps were recorded and logged, with data being sampled during ten cycles.

3.5 VAWT Experiment

The VAWT experiment was designed to emulate the incidence time histories encountered by the blade of a vertical-axis wind turbine. A computer algorithm, coded in FORTRAN 77, has been developed at the University of Reading to calculate the blade's angle of attack as a function of its azimuth position. The program can use both single and multiple streamtube models¹³ based on SANDIA¹⁴ data for the NACA series of aerofoil characteristics.

At low tip-speed ratios the time history for the single streamtube model is a skewed sine function, but this tends toward a true sine as the tip-speed ratio is increased. The upwind (positive) and downwind (negative) sections of each cycle attain identical peak values of incidence. Tip-speed ratio and amplitude are related as follows:

	and the second se
TSR	Amplitude
6.00	5.4°
4.00	9.9°
3.50	12.2°
3.25	13.8°
2.80	17.4°
2.33	22.6°
1.75	32.8°

The AMSTRAD function generator reproduced the angle of attack histories based upon the NACA 0015 aerofoil's characteristics. Data acquisition was performed in an identical manner to that for sinusoidal tests.

3.6 Procedure

Before each individual set of tests, the tunnel was shut down and the air flow allowed to cease before the transducer offsets were logged. Immediately after these values were recorded, the appropriate data acquisition routine was initiated whilst the tunnel was brought up to speed and thence data gathered as per the software prompts. The tunnel was then shut down, offsets logged again and further tests were performed in the manner described above.

3.7 Roughness Transition Strips

A number of the experiments were repeated with graded sand deposited at the aerofoil's leading edge. It was intended that this should trip the boundary layer in the leading-edge region. A direct comparison can be made between tests with and those without these roughness transition strips.

3.8 Data Presentation

All data collected by the data acquisition routines were stored in unformatted form on magnetic tape. A library of programs (coded in FORTRAN 77) is available for the reduction, presentation and analysis of the data on a DEC MICROVAX 3400. By applying offsets, gains and calibrations, the data reduction programs convert the cycles of raw data into averaged or unaveraged non-dimensional pressure coefficients. As described by Leitch and Galbraith¹⁵, these data are stored on the University of Glasgow's aerofoil database. The airloads are determined by suitably integrating the pressure coefficient values.

4 RESULTS AND DISCUSSION

4.1 Tunnel Performance

Assessment of the quality of the data can only be made with a clear insight of the tunnel effects. Unfortunately the tunnel performance was such that, for the time scales of the model motion, it was not possible to hold the dynamic pressure in the working section constant whilst altering the blockage due to the pitching of the aerofoil. During the static tests (i.e. $\mathbf{k}=0.0$ and $\mathbf{r}=0.0$), this variation was as illustrated in **Figure 5**, where it can be seen that there was approximately a 30% reduction in dynamic pressure as the angle of attack was increased from 0° to 30°. As illustrated in **Figures 6** and **7**, this reduction in dynamic pressure decreased as reduced frequency increased. Figure 8 reveals that, during ramps, there was a drastic reduction and subsequent unsteadiness in the dynamic pressure during a test. The model was pitched to an incidence of 40° so that uniform ramp conditions existed at stall. Once the aerofoil had stalled, however, all significant data had already been collected and the corresponding dynamic pressure reduction was only in the region of 10%. The subsequent data are of little relevance to the current work and is presented merely for completeness.

4.2 Averaging of the Data

The main data in this report are the average of a number of cycles. Individual cycles are presented in **Figures 9** and **10** where it may be seen that, whilst minor random differences do exist from cycle to cycle, the salient features are highlighted by the averaging process. In addition, the sweep at which any event occurred did not vary. Therefore the given data may be considered as typical of aerofoil performance during any given individual cycle. This is particularly relevant when considering the detailed flow phenomena of separation and reattachment.

4.3 Test Data

The test data are grouped for each motion type with compact details of the specific tests listed in **Tables 2** to **7**.

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TABLE 1 : NACA 0015 AEROFOIL PROFILE AND COORDINATES



Coordinates in %Chord

Upper S	Surface	Lower	Surface
Station	Ordinate	Station	Ordinate
0.000	0.000	0.000	0.000
0.137	0.811	0.137	-0.811
0.548	1.596	0.548	-1.596
1.231	2.350	1.231	-2.350
2.185	3.073	2.185	-3.073
3.407	3.759	3.407	-3.759
4.894	4.403	4.894	-4.403
6.642	5.001	6.642	-5.001
8.645	5.547	8.645	-5.547
10.899	6.035	10.899	-6.035
13.397	6.460	13.397	-6.460
16.133	6.817	16.133	-6.817
19.098	7.103	19.098	-7.103
22.285	7.314	22.285	-7.314
25.686	7.447	25.686	-7.447
29.289	7.501	29.289	-7.501
33.087	7.476	33.087	-7.476
37.068	7.373	37.068	-7.373
41.221	7.194	41.221	-7.194
45.536	6.941	45.536	-6.941
50.000	6.618	50.000	-6.618
54.601	6.227	54.601	-6.227
59.326	5.773	59.326	-5.773
64.163	5.259	64.163	-5.259
69.098	4.689	69.098	-4.689
74.118	4.064	74.118	-4.064
79.209	3.388	79.209	-3.388
84.357	2.660	84.357	-2.660
89.547	1.880	89.547	-1.880
94.766	1.047	94.766	-1.047
100.000	0.158	100.000	-0.158

TABLE 2 : DETAILS OF STATIC TESTS

TABLE 2.1 : SUMMARY OF STATIC TESTS (nominal)

Reynolds Number	1.0x10 ⁶	1.5x10 ⁶	2.0x10 ⁶
Angle of Attack		-2° to 30°	

(all permutations)

TABLE 2.2 : LIST OF STATIC TESTS (actual)

Run	Start	Sweep	Reynolds
Number	C	Ô	No. x 10 ⁻⁶
00411	-2	32	1.04
00421	-2	32	1.01
00431	-2	32	0.98
00441	30	32	0.90
00451	-2	32	1.01
00461	-2	32	0.78
00471	-2	32	0.77
00481	-2	32	0.77
00491	-2	32	1.06
00511	-2	32	0.95
00531	-2	32	0.85
00551	-2	32	0.76
00571	-2	32	0.66
00591	-2	32	0.55
00611	-2	32	0.83
00781	-2	32	0.83
00831	-2	32	1.08
01111	-2	32	2.15
01211	-2	32	1.69
01251	-2	32	1.58
01381	-2	32	1.58
01661	-2	32	1.57
01721	-2	32	2.13
*801971	-2	32	2.20
*801981	-2	32	1.77

*experiments with roughness transition strips

TABLE 3 : DETAILS OF UNSTEADY STATIC TESTS

TABLE 3.1 : SUMMARY OF UNSTEADY STATIC TESTS (nominal)

Incidence Range	0° - 35°
Sampling Frequency	500Hz
Reynolds Number	2.0×10^{6}

TABLE 3.2 : LIST OF UNSTEADY STATIC TESTS (actual)

1.1.1.1.1.1.1.1	Angle of	Sampling	Reynolds
Run	Attack	Frequency	Number
Number	(°)	(Hz)	x 10 ⁻⁶
40841	0	500	1.06
40851	1	500	1.06
40861	2	500	1.05
40871	3	500	1.04
40881	4	500	1.04
40891	5	500	1.04
40901	6	500	1.04
40911	7	500	1.04
40921	8	500	1.04
40931	9	500	1.03
40941	10	500	1.03
40951	11	500	1.03
40961	12	500	1.03
40971	13	500	2.06
40981	14	500	2.08
40991	15	500	2.07
41001	16	500	2.06
41011	17	500	2.07
41021	18	500	2.07
41031	19	500	2.06
41041	20	500	2.06
41051	21	500	2.06
41061	22	500	2.05
41071	23	500	2.05
41081	24	500	2.04
41091	25	500	2.04
41101	26	500	2.04
41121	27	500	2.12
41131	28	500	2.11
41141	29	500	2.10
41151	30	500	2.10
41161	31	500	2.09
41171	32	500	2.09
41181	33	500	2.09
41191	34	500	2.08
41201	35	500	2.08

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TABLE 4 : DETAILS OF RAMP UP TESTS

TABLE 4.1 : SUMMARY OF FAST RAMPS (nominal)

Starting Angle	-1 ^o											
Finishing Angle	40°											
Pitch Rate (°s ⁻¹)	180	210	240	270	300	335	370	400	435	470	500	530
Reynolds Number	0.8×10^6 1.0×10^6											

(all permutations, with and without roughness transition strips)

TABLE 4.2 : SUMMARY OF SLOW RAMPS (nominal)

Starting Angle		-1 [°]								145				
Finishing Angle	40°													
Pitch Rate (°s ⁻¹)	2	4	8	12	16	20	30	40	60	80	100	120	150	160
Reynolds Number		0.8x10 ⁶												

(all permutations, without roughness transition strips)

TABLE 4.3 : LIST OF RAMP UP TESTS (actual)

Run	Start	Arc	Pitch Rate	Reduced	Reynolds
Number	(°)	(°)	$(^{o}s^{-1})$	Pitch Rate	No. x 10 ⁻⁶
20023	-1	41	1.9	0.0001	0.78
20033	-1	41	3.8	0.0002	0.78
20043	-1	41	7.7	0.0004	0.77
20053	-1	41	11.9	0.0007	0.77
20063	-1	41	16.0	0.0009	0.77
20073	-1	41	20.4	0.0011	0.77
20083	-1	41	37.6	0.0021	0.76
20093	-1	41	79.8	0.0044	0.76
20103	-1	41	123.4	0.0069	0.76
20113	-1	41	168.3	0.0094	0.76
20123	-1	41	197.9	0.0110	0.76
20133	-1	41	252.5	0.0140	0.76
20143	-1	41	284.2	0.0159	0.77
20153	-1	41	320.1	0.0179	0.76
20163	-1	41	355.2	0.0198	0.76
20173	-1	41	366.3	0.0204	0.76
20183	-1	41	444.8	0.0249	0.76
20194	-1	41	442.3	0.0247	0.76
21221	-1	40	463.9	0.0257	0.80
21231	-1	40	501.4	0.0278	0.80
21241	-1	41	534.5	0.0297	0.80
21261	-1	41	16.6	0.0009	0.80
21271	-1	40	20.5	0.0011	0.80
21281	-1	41	29.2	0.0016	0.79
21291	-1	40	40.7	0.0023	0.79
21301	-1	41	57.8	0.0032	0.80

TABLE 4.3 : LIST OF RAMP UP TESTS (concluded)

Run	Start	Arc	Pitch Rate	Reduced	Reynolds
Number	(°)	(9)	$(^{\circ}s^{-1})$	Pitch Rate	No. $\times 10^{-6}$
21311	-1	41	80.6	0.0045	0.79
21321	-1	40	98.3	0.0054	0.80
21331	-1	40	124.2	0.0069	0.80
21341	-1	40	151.8	0.0083	0.81
21351	-1	40	183.5	0.0100	0.80
21361	-1	40	213.1	0.0117	0.80
21371	-1	40	242.5	0.0132	0.80
21391	-1	41	274.4	0.0156	0.79
21401	-1	40	309.2	0.0175	0.79
21411	-1	40	334.3	0.0188	0.79
21421	-1	40	366.5	0.0205	0.79
21431	-1	40	398.8	0.0225	0.79
21441	-1	40	434.3	0.0242	0.80
21/31	-1	40	181.6	0.0074	1.06
21/41	-1	40	211.3	0.0086	1.05
21/51	-1	40	242.7	0.0098	1.04
21701	-1	40	272.2	0.0110	1.04
21771	-1	40	304.5	0.0125	1.03
21701	-1	40	368 7	0.0154	1.04
21801	-1	40	399 5	0.0151	1.01
21811	-1	40	432.2	0.0102	1.02
21821	-1	40	469.9	0.0192	1.00
21831	-1	40	498.8	0.0204	1.00
21841	-1	40	529.4	0.0215	1.00
*821991	-1	40	181.4	0.0092	0.88
*822001	-1	40	213.0	0.0108	0.88
*822011	-1	40	241.7	0.0123	0.87
*822021	-1	40	271.4	0.0138	0.87
*822031	-1	40	306.8	0.0155	0.87
*822041	-1	40	336.7	0.0170	0.87
*822051	-1	40	368.1	0.0186	0.86
*822061	-1	40	399.1	0.0201	0.86
*822071	-1	40	436.1	0.0221	0.86
*822081	-1	40	474.0	0.0239	0.86
*822091	-1	40	500.6	0.0253	0.85
*822101	-1	40	534.8	0.0271	0.85
*822231	-1	41	181.2	0.0075	1 10
*822241	-1	41	211.3	0.0087	1.09
*822251	-1	41	2417	0.0099	1.09
*822261	-1	41	272 1	0.0111	1.09
*822201	-1	41	272 1	0.0111	1.00
*822281	-1	41	335.2	0.0138	1.05
*822201	-1	41	368 2	0.0150	1.05
*822301	-1	41	398.9	0.0163	1.05
*822211	_1	41	433.0	0.0176	1.04
*022311	_1	41	470.6	0.0100	1.04
*822221		41	106.6	0.0190	1.03
*822241	1	41	530 /	0.0202	1.05
022341	-1	-1	550.4	0.0215	1.04

*experiments with roughness transition strips

TABLE 5 : DETAILS OF RAMP DOWN TESTS

TABLE 5.1 : SUMMARY OF RAMPS FROM 40° TO -1° AT A REYNOLDS NUMBER OF 0.8x10⁶ (nominal)

Starting Angle		40°										
Finishing Angle		-1°										
Pitch Rate	-2	-4	-8	-12	-16	-20	-40	-80	-120	-160	-200	-230
(°s ⁻¹)		-260	-290	-320	-360	-400	-440	-470	-500	-500		
Reynolds Number		0.8x10 ⁶										

(all permutations, without roughness transition strips)

TABLE 5.2 : SUMMARY OF RAMPS FROM 40° TO -1° AT A REDUCED PITCH RATE OF -0.02 (nominal)

Starting Angle	40°									
Finishing Angle		-1°								
Reduced Pitch Rate	1224.	-0.02								
Reynolds Number	$0.6x10^{6} 0.7x10^{6} 0.8x10^{6} 0.9x10^{6} 1.0x10^{6} 1.1x10^{6}$									

(all permutations, without roughness transition strips)

TABLE 5.3 : SUMMARY OF RAMPS TO -10° (nominal)

Starting Angle	40°	35°	30°	28°	26°	24°	22°	20°	18°	16°	14°	12°
Finishing Angle		-10°										
Pitch Rate		-450°s ⁻¹										
Reynolds Number		0.8x10 ⁶										

(all permutations, without roughness transition strips)

TABLE 5.4 : SUMMARY OF RAMPS FROM 40° TO -10° (nominal)

Starting Angle		40°								
Finishing Angle		-10°								
Pitch Rate (°s ⁻¹)	-230	-230 -270 -310 -340 -380 -410 -440 -470 -500 -550 -600							-650	
Reynolds Number	256	0.8×10^6 1.0×10^6								

(all permutations, with and without roughness transition strips)

TABLE 5.5 : LIST OF RAMP DOWN TESTS (actual)

Run	Start	Am	Pitch Rate	Reduced	Reynolde
Number	(9)	(0)	$(^{\circ}e^{-1})$	Pitch Rate	No $\times 10^{-6}$
30223	40	41	19	0.0001	0.76
30223	40	41	-1.8	-0.0001	0.76
30233	40	41	-4.0	-0.0002	0.76
30243	40	41	-7.5	-0.0004	0.76
30255	40	41	-11.5	-0.0000	0.76
30203	40	-41	-13.0	-0.0009	0.76
30275	40	-41	-19.5	-0.0011	0.76
30283	40	-41	-39.1	-0.0022	0.75
30293	40	-41	-70.0	-0.0043	0.75
30303	40	-41	-120.2	-0.0067	0.78
30313	40	-41	-159.0	-0.0089	0.78
30323	40	-41	-200.2	-0.0112	0.78
30333	40	-41	-230.6	-0.0129	0.77
30343	40	-41	-255.9	-0.0143	0.77
30353	40	-41	-288.6	-0.0161	0.77
30363	40	-41	-326.8	-0.0183	0.77
30373	40	-41	-367.8	-0.0205	0.77
30383	40	-41	-404.4	-0.0230	0.75
30394	40	-41	-438.1	-0.0249	0.75
30501	40	-41	-545.0	-0.0224	1.05
30521	40	-41	-490.3	-0.0218	0.95
30541	40	-41	-445.0	-0.0220	0.85
30561	40	-41	-382.8	-0.0219	0.76
30581	40	-41	-323.6	-0.0213	0.65
30601	40	-41	-262.0	-0.0207	0.55
30621	40	-50	-541.3	-0.0288	0.82
30631	35	-45	-408.6	-0.0217	0.82
30641	30	-40	-430.3	-0.0228	0.81
30651	28	-38	-425.3	-0.0225	0.81
30661	26	-36	-433.3	-0.0231	0.81
30671	24	-34	-448.3	-0.0239	0.81
30681	22	-32	-445.2	-0.0237	0.80
30691	20	-30	-453.7	-0.0241	0.80
30791	18	-28	-445.5	-0.0238	0.82
30801	16	-26	-430.9	-0.0230	0.82
30811	14	-24	-426.3	-0.0228	0.82
30821	12	-22	-410.6	-0.0219	0.82
31451	40	-50	-21.7	-0.0014	0.71
31461	40	-50	-26.7	-0.0017	0.72
31471	40	-50	-38.0	-0.0024	0.71
31481	40	-50	-53.0	-0.0034	0.70
31491	40	-50	-75.6	-0.0047	0.72
31501	40	-50	-103.7	-0.0065	0.70
31511	40	-50	-123.9	-0.0077	0.71
31521	40	-50	-153.7	-0.0095	0.71
31531	40	-50	-186.1	-0.0116	0.70
31541	40	-50	-230.4	-0.0140	0.72
31551	40	-50	-276.5	-0.0174	0.69
31561	40	-50	-302.1	-0.0189	0.70
31571	40	-50	-342.5	-0.0214	0.70
31581	40	-50	-379.5	-0.0260	0.67
31591	40	-50	-409.7	-0.0278	0.67

TABLE 5.5 : LIST OF RAMP DOWN TESTS (concluded)

Run	Start	Arc	Pitch Rate	Reduced	Reynolds
Number	(\circ)	(°)	(°s ⁻¹)	Pitch Rate	No. x 10 ⁻⁶
31601	40	-50	-447.9	-0.0306	0.66
31611	40	-50	-476.8	-0.0322	0.67
31621	40	-50	-507.6	-0.0341	0.67
31631	40	-50	-549.5	-0.0367	0.67
31641	40	-50	-595.0	-0.0394	0.67
31651	40	-50	-645.4	-0.0432	0.67
31851	40	-50	-231.4	-0.0108	0.91
31861	40	-50	-272.2	-0.0126	0.91
31871	40	-50	-311.4	-0.0146	0.90
31881	40	-50	-344.6	-0.0162	0.88
31891	40	-50	-379.2	-0.0180	0.87
31901	40	-50	-411.3	-0.0192	0.88
31911	40	-50	-420.0	-0.0199	0.80
31921	40	-50	-4/8.1	-0.0225	0.88
31931	40	-50	-510.5	-0.0235	0.89
31941	40	-50	-338.8	-0.0262	0.87
31951	40	-50	-007.7	-0.0201	0.87
*020111	40	-50	-034.9	-0.0300	0.87
*020101	40	50	-232.8	-0.0154	0.73
*022121	40	50	-2/5.4	-0.0134	0.77
*832131	40	50	-309.1	-0.01/0	0.76
*832141	40	50	-341.9	-0.0197	0.74
*832151	40	50	-311.2	-0.0215	0.75
*832161	40	50	-412.4	-0.0240	0.74
*832171	40	50	-440.5	-0.0252	0.75
*832181	40	50	-473.2	-0.0272	0.74
*832191	40	50	-506.7	-0.0284	0.76
*832201	40	50	-559.2	-0.0321	0.74
*832211	40	50	-606.4	-0.0351	0.73
*832221	40	50	-658.9	-0.0377	0.74
*832351	40	50	-229.9	-0.0108	0.90
*832361	40	50	-273.3	-0.0128	0.89
*832371	41	50	-312.8	-0.0149	0.87
*832381	40	50	-345.1	-0.0161	0.89
*832391	40	50	-346.2	-0.0162	0.88
*832401	40	50	-394.8	-0.0184	0.88
*832411	40	50	-445.0	-0.0208	0.88
*922421	40	50	-477 9	-0.0223	0.88
*822421	40	50	-509 3	-0.0240	0.86
*022431	40	50	-562.2	-0.0257	0.80
*020451	40	50	614.0	0.0237	0.87
832451	40	50	-014.0	-0.0280	0.87
*832461	40	50	-661.9	-0.0308	0.87

*experiments with roughness transition strips

TABLE 6 : DETAILS OF SINUSOIDAL TESTS

TABLE 6.1 : SUMMARY OF SINUSOIDAL TESTS (nominal)

Mean Angle	4°	6°	8°	15°	16°	17°	18°	20°
Amplitude	10°							
Reduced Frequency	0.175							
Reynolds Number	0.8x10 ⁶						1.	

(all permutations, without roughness transition strips)

TABLE 6.2 : SUMMARY OF SINUSOIDAL TESTS (actual)

Run	Mean	Amp'ude	Reduced	Reynolds	
Number	()	()	Frequency	No. x 10 °	
10702	4	10	0.168	0.83	
10712	6	10	0.167	0.83	
10722	8	10	0.167	0.82	
10732	15	10	0.167	0.82	
10742	16	10	0.167	0.82	
10752	17	10	0.167	0.82	
10762	18	10	0.167	0.82	
10772	20	10	0.167	0.82	

TABLE 7 : DETAILS OF VAWT TESTS

TABLE 7.1 : SUMMARY OF VAWT TESTS (nominal)

Mean Angle	0°						
TSR	1.75						
Reduced Frequency	0.040	0.080	0.100	0.125	0.150		
Reynolds Number	1.5x10 ⁶						

(all permutations, without roughness transition strips)

TABLE 7.2 : SUMMARY OF VAWT TESTS (actual)

Run Number	Mean (°)	TSR	Reduced Frequency	Reynolds No. x 10 ⁶
51671	0	1.75	0.145	1.58
51681	0	1.75	0.118	1.57
51691	0	1.75	0.097	1.57
51701	0	1.75	0.076	1.58
51711	0	1.75	0.037	1.58



FIGURE 1 : GLASGOW UNIVERSITY'S DYNAMIC STALL RIG





7ft X 5ft Jin WIND TUNNEL



FIGURE 3 : PRESSURE TRASDUCER LOCATIONS FOR THE NACA (0015.















FIGURE 9: EFFECT OF AVERAGING ON THE NORMAL FORCE AND PITCHING MOMENT FOR OSCILLATORY TESTS.



FIGURE 10: TYPICAL UNAVERAGED DATA FOR RAMP TESTS.

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PRESSURE DATA FROM

STATIC EXPERIMENTS



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DYNAMIC PRESSURE = 1088.11 Nm⁻² REYNOLDS NUMBER = 771138.

NUMBER OF CYCLES = 1



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PRESSURE DATA FROM

UNSTEADY STATIC EXPERIMENTS





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DYNAMIC PRESSURE = 2091.56 Nm^{-2} MOTION TYPE: UNSTEADY STATIC RUN REFERENCE NUMBER: 40971 REYNOLDS NUMBER = 1029687. NUMBER OF CYCLES = 1

SAMPLING FREQUENCY = 500.00 Hz. ANGLE OF ATTACK = 13.0 AIR TEMPERATURE = 35.3°C DATE OF TEST: 26/8/91 MACH NUMBER = 0.171



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Cps at LE, TE, 30%





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Cps at LE, TE, 30%

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DYNAMIC CHARACTERISTICS FOR THE NACA 0015 - half chord

DATE OF TEST: 26/8/91

RUN REFERENCE NUMBER: 41071 REYNOLDS NUMBER = 1024578.

NUMBER OF CYCLES = 1

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-CP 2

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PRESSURE DATA FROM

RAMP UP EXPERIMENTS







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DYNAMIC PRESSURE = 1329.17 Nm⁻² RUN REFERENCE NUMBER: 822001 REYNOLDS NUMBER = 875670. START ANGLE = -1.00° NUMBER OF CYCLES = 5 MOTION TYPE: RAMP UP RAMP ARC = 40.000°

AVERAGED DATA OF 5 CYCLES

SAMPLING FREQUENCY = 5208.00 Hz. LINEAR PITCH RATE = 213.03°S⁻¹ REDUCED PITCH RATE = 0.01082 AIR TEMPERATURE = 16.8°C DATE OF TEST: 1/10/91 MACH NUMBER = 0.138





ANGLE OF ATTACK (alpha) Non-dimensional time (txV/c)





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Cps at LE, TE, 30%


-Cb



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-Cp



















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PRESSURE DATA FROM

RAMP DOWN EXPERIMENTS





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-21





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DYNAMIC PRESSURE = 1509.30 Nm^{-2} RUN REFERENCE NUMBER: 832461 REYNOLDS NUMBER = 867816. MOTION TYPE: RAMP DOWN START ANGLE = 40.00° NUMBER OF CYCLES = 5 RAMP ARC = -50.000°

5 CYCLES

AVERAGED DATA OF

-40 Non-dimensional time (txV/c) -2 Non-dimensional time (txV/c) -20-ANGLE SAMPLING FREQUENCY = 12760.00 Hz. LINEAR PITCH RATE = -661.89°S⁻¹ REDUCED PITCH RATE = -0.03085 AIR TEMPERATURE = 36.1°C DATE OF TEST: 3/10/91 MACH NUMBER = 0.146



-8.94

ANGLE OF ATTACK (alpha) Non-dimensional time (txV/c)



10.4

2.

15.5

18.1

20.75

23.30

57

25. 27.77 29.63 alpha







246

-0.3L

-0.31

14 16

P.

246

Cps at LE, TE, 30%

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PRESSURE DATA FROM

SINUSOIDAL EXPERIMENTS





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-Cp 2

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PRESSURE DATA FROM

VAWT EXPERIMENTS





Q Q 1

- 00 N



TOT



-2 PO

N 2 4 0 0 7 8

dy-



40,

OSCILLATION FREQUENCY = 0.930 Hz. DYNAMIC PRESSURE = 1107.43 Nm⁻² RUN REFERENCE NUMBER: 51711 MOTION TYPE: VAWT FUNCTION REYNOLDS NUMBER = 1579272. NUMBER OF CYCLES = 10 0.00° MEAN ANGLE =

AVERAGED DATA OF 10 CYCLES

SAMPLING FREQUENCY = 119.03 Hz. REDUCED FREQUENCY = 0.037 AIR TEMPERATURE = 23.8°C DATE OF TEST: 5/9/91 MACH NUMBER = 0.124**AMPLITUDE = 32.80^{\circ}**





26.33 28.30 15.90

-0.28

-16.99

29.83 22.72 1.91 -6.56/

-NW + 500 78

d'

alpha

0.6 0.8 x/c

og v













ct -0.2

-0.1

-0.3

171

