A STUDY ON THE FAILURE PREDICTION OF COMPOSITE LAMINATES IN BENDING

Jinsung Kim, Korea Aerospace University Jinsungkim83@naver.com Sooyong Lee, Korea Aerospace University Jinho Roh, Korea Aerospace University Jaesung Bae, Korea Aerospace University

Key Words: Composite Laminate, Failure, Non-linear Element Analysis, Bending

Failure prediction for composite materials under given loading conditions is important for efficient design in structural applications. Over the past several decades, there are numerous failure criteria proposed to more accurately predict the failure composite laminates. A lot of research was conducted to evaluate and validate the failure prediction capability for failure criteria. The most failure criteria are studied for in-plane loading conditions. Mechanical behavior of composite laminates varies depending on the loading conditions. Even if failure criterion is accurate under the in-plane loads, it cannot be accurate for out-of-plane loads such as bending. In many industrial structures, composite laminates is under out-of-plane load as well as in-plane loads. For the structural stability of the composite structures, it is important to accurately predict failure of composite laminates under bending.

In this study, the failure prediction of composite laminates under bending is investigated. The non-linear finite element analysis using Arc-length method is performed. 2D strain-based interactive failure theory [1] that is more accurately final failure of composite laminate under multi-axial loading is applied to predict the final failure of composite laminate under multi-axial loading is applied to predict the final failure of composite laminate under multi-axial loading is applied to predict the final failure of composite laminates under to compare the accuracy of the failure predictions, a 3-point bending test are performed for un-symmetric cross-ply [0/90]₈ and quasi-isotropic [0/±45/90]_{2s} composite laminates. Also, it is compared with the other failure criteria such as maximum strain, maximum stress and Tsai-Wu theories. Finally, the predicted results using 2D strain-based interactive failure theory more agree well with the experiment than other failure theories.

Acknowledgements

This work was supported under the framework of Aerospace Technology Development Program (No. 10074270, Development of Manufacturing Core Technology for 3-Dimnesional Woven Integrated Composite Wing Structure of 5,000 Pound VLJ Aircraft) funded by the Ministry of Trade, Industry & Energy (MOTIE, Korea) This work was supported by the New & Renewable Energy Core Technology Program of the Korea Institute of Energy Technology Evaluation and Planning (KETEP) granted financial resource from the Ministry of Trade, Industry & Energy, Republic of Korea. (No. 20143030021130)

References

[1] S. Y. Lee and J. H. Roh, "Two-dimensional strain-based interactive failure theory for multidirectional composite laminates," *Composite Part B: Engineering*, vol. 69, pp.69-75, 2015.